COMPONENT MAINTENANCE MANUAL WITH ILLUSTRATED PARTS LIST

FWD GALLEY DOOR TORQUE TUBE ASSEMBLY

PART NUMBER 65-45871–102, –113, –142, –88

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COMPONENT MAINTENANCE MANUAL

Revision No. 14
Jul 01/2009

To: All holders of FWD GALLEY DOOR TORQUE TUBE ASSEMBLY 52-46-22.

Attached is the current revision to this COMPONENT MAINTENANCE MANUAL

The COMPONENT MAINTENANCE MANUAL is furnished either as a printed manual, on microfilm, or digital products, or any combination of the three. This revision replaces all previous microfilm cartridges or digital products. All microfilm and digital products are reissued with all obsolete data deleted and all updated pages added.

For printed manuals, changes are indicated on the List of Effective Pages (LEP). The pages which are revised will be identified on the LEP by an R (Revised), A (Added), O (Overflow, i.e. changes to the document structure and/or page layout), or D (Deleted). Each page in the LEP is identified by Chapter-Section-Subject number, page number and page date.

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Location of Change	Description of Change
	NO HIGHLIGHTS

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COMPONENT MAINTENANCE MANUAL

A = Added, R = Revised, D = Deleted, O = Overflow

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COMPONENT MAINTENANCE MANUAL

TEMPORARY REVISION AND SERVICE BULLETIN RECORD

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVE	DATE OF INCORPORATION INTO MANUAL
52-1094		PRR 34272	JUN 05/88

All revisions to this manual will be accompanied by transmittal sheet bearing the revision number. Enter the revision number in numerical order, together with the revision date, the date filed and the initials of the person filing.

Revision		Filed	
Number	Date	Date	Initials

Revision		Filed	
Number	Date	Date	Initials

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All temporary revisions to this manual will be accompanied by a cover sheet bearing the temporary revision number. Enter the temporary revision number in numerical order, together with the temporary revision date, the date the temporary revision is inserted and the initials of the person filing.
When the temporary revision is incorporated or cancelled, and the pages are removed, enter the date the pages are removed and the initials of the person who removed the temporary revision.

Temporary Revision		Inserted		Removed	
Number	Date	Date	Initials	Date	Initials

Temporary Revision		Inserted		Removed	
Date	Initials	Number	Date	Date	Initials

Temporary Revision		Inserted		Removed		Temporary Revision		Inserted		Removed	
Number	Date	Date	Initials	Date	Initials	Date	Initials	Number	Date	Date	Initials

INTRODUCTION

1. General

A. The instructions in this manual supply the data necessary to do the maintenance functions together with the test, fault isolation, repair, and replacement of the defective parts.

B. This manual is divided into different parts:

(1) Title Page

(2) Transmittal Letter

(3) Highlights

(4) List of Effective Pages

(5) Table of Contents

(6) Temporary Revision & Service Bulletin Record

(7) Record of Revisions

(8) Record of Temporary Revisions

(9) Introduction

(10) Procedures & IPL Sections

C. Components that can be repaired have a different repair number for each specified repair. To find the repair number location of a component, look in the Repair-General procedure at the beginning of the REPAIR section. The Repair-General procedure also has an explanation of the True Position Dimension symbols used.

D. All dimensions, measures, quantities and weights included are in English units. When metric equivalents are given they will be in the parentheses that follow the English units.

E. The introduction to the Illustrated Parts List (IPL) shows how the IPL data is used.

F. Design changes, optional parts, configuration differences and Service Bulletin modifications may cause different part numbers. These part numbers are identified in the IPL with an alphabetical letter which is added to the end of the basic item number. This new item number is referred to as an alpha-variant. Throughout the manual, IPL basic item number references also apply to alpha-variants unless shown differently.

G. The tool reference numbers found in the individual procedures and in the Special Tools, Fixtures, and Equipment section are used to identify if a tool is a standard tool (STD-XXXX), a commercial tool (COM-XXXX), or a Special Tool (SPL-XXXX). This reference number is also used to distinguish between tools with similar names in the same procedure. These reference numbers are for use in the documentation only. They are not to be used for ordering tools.

FORWARD GALLEY DOOR TORQUE TUBE ASSEMBLY - DESCRIPTION AND OPERATION

1. Description

A. The forward galley door torque tube assembly is installed in the fuselage of the airplane. The torque tube assembly includes a torque tube, two sleeves, two hinge link pins, and related fasteners.

2. Operation

A. The torque tube assembly helps to open and close the forward galley door.

3. Leading Particulars (Approximate)

A. Length – 25 inches

B. Diameter – 1.20 inches

C. Weight – 5 pounds

COMPONENT MAINTENANCE MANUAL

TESTING AND FAULT ISOLATION

(NOT APPLICABLE)

DISASSEMBLY

1. General

A. This procedure has the data necessary to disassemble the forward galley door torque tube assembly.

B. Disassemble this component sufficiently to isolate the defects, do the necessary repairs, and put the component back to a serviceable condition.

C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

D. Refer to IPL Figure 1 for item numbers.

2. Disassembly

A. References

Reference	Title
SOPM 20-50-01	BOLT AND NUT INSTALLATION

B. Procedure

(1) Use standard industry practices and these steps.

NOTE: For bolt and nut installation, refer to SOPM 20-50-01.

(a) Remove bolt (10), washer (15), nut (20), and pin (5).

(b) Remove bolt (30), washer (35), nut (40), and sleeve (25).

(c) Remove bolt (50), washer (55), nut (60), and pin (45).

(d) Remove bolt (70), washer (75), nut (80), and sleeve (65).

(NOT APPLICABLE)

CHECK

1. General

A. This procedure has the data necessary to find defects in the material of the specified parts.

B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

2. Check

A. References

Reference	Title
SOPM 20-20-01	MAGNETIC PARTICLE INSPECTION

B. Procedure

(1) Examine all parts for defects by standard industry practices.

(2) Magnetic particle check (SOPM 20-20-01) – pins (5, 45), sleeves (25, 65), torque tube (85).

REPAIR

1. **Content**

   A. Repair, refinish and replacement procedures are included in separate repair section as follows:

Table 601:

P/N	NAME	REPAIR
66-14525	HINGE LINK PIN	1-1
66-14526	HINGE LINK PIN	2-1
66-14537	TORQUE TUBE	3-1
66-23571	TORQUE TUBE SLEEVE	4-1

2. **Standard Practices**

   A. Refer to the following standard practices, as applicable, for details of procedures in individual repairs.

   - SOPM 20-30-02 Stripping of Protective Finishes
   - SOPM 20-41-01 Decoding Table for Boeing Finish Codes
   - SOPM 20-41-02 Application of Chemical and Solvent Resistant Finishes
   - SOPM 20-42-05 Bright Cadmium Plating
   - SOPM 20-60-02 Finishing Materials

3. **Materials and Equipment**

   NOTE: Equivalent substitutes can be used.

   A. primer, C00259 (SOPM 20-60-02)

4. **Dimensioning Symbols**

   A. Standard true position dimensioning symbols used in applicable repair procedures are shown in REPAIR-GENERAL, Figure 601.

—	STRAIGHTNESS
▱	FLATNESS
⊥	PERPENDICULARITY (OR SQUARENESS)
//	PARALLELISM
○	ROUNDNESS
⌭	CYLINDRICITY
⌒	PROFILE OF A LINE
⌓	PROFILE OF A SURFACE
◎	CONCENTRICITY
⌯	SYMMETRY
∠	ANGULARITY
↗	RUNOUT
↗↗	TOTAL RUNOUT
⌴	COUNTERBORE OR SPOTFACE
∨	COUNTERSINK
⌖	THEORETICAL EXACT POSITION OF A FEATURE (TRUE POSITION)
∅	DIAMETER
S ∅	SPHERICAL DIAMETER
R	RADIUS
SR	SPHERICAL RADIUS
()	REFERENCE
BASIC (BSC) OR [DIM]	A THEORETICALLY EXACT DIMENSION USED TO DESCRIBE SIZE, SHAPE OR LOCATION OF A FEATURE. FROM THIS FEATURE PERMISSIBLE VARIATIONS ARE ESTABLISHED BY TOLERANCES ON OTHER DIMENSIONS OR NOTES.
[-A-]	DATUM
Ⓜ	MAXIMUM MATERIAL CONDITION (MMC)
Ⓛ	LEAST MATERIAL CONDITION (LMC)
Ⓢ	REGARDLESS OF FEATURE SIZE (RFS)
Ⓟ	PROJECTED TOLERANCE ZONE
FIM	FULL INDICATOR MOVEMENT

EXAMPLES

— | 0.002	STRAIGHT WITHIN 0.002
⊥ | 0.002 | B	PERPENDICULAR TO DATUM B WITHIN 0.002
// | 0.002 | A	PARALLEL TO DATUM A WITHIN 0.002
○ | 0.002	ROUND WITHIN 0.002
⌭ | 0.010	CYLINDRICAL SURFACE MUST LIE BETWEEN TWO CONCENTRIC CYLINDERS, ONE OF WHICH HAS A RADIUS 0.010 INCH GREATER THAN THE OTHER
⌒ | 0.006 | A	EACH LINE ELEMENT OF THE SURFACE AT ANY CROSS SECTION MUST LIE BETWEEN TWO PROFILE BOUNDARIES 0.006 INCH APART RELATIVE TO DATUM A
⌓ | 0.020 | A	SURFACES MUST LIE WITHIN PARALLEL BOUNDARIES 0.020 INCH APART AND EQUALLY DISPOSED ABOUT TRUE PROFILE
◎ | ∅ 0.0005 | C	CONCENTRIC TO DATUM C WITHIN 0.0005 DIAMETER
⌯ | 0.010 | A	SYMMETRICAL WITH DATUM A WITHIN 0.010
∠ | 0.005 | A	ANGULAR TOLERANCE 0.005 WITH DATUM A
⌖ | ∅ 0.002 Ⓢ | B	LOCATED AT TRUE POSITION WITHIN 0.002 DIA RELATIVE TO DATUM B, REGARDLESS OF FEATURE SIZE
⊥ | ∅ 0.010 Ⓜ | A 0.510 Ⓟ	AXIS IS TOTALLY WITHIN A CYLINDER OF 0.010 INCH DIAMETER, PERPENDICULAR TO DATUM A, AND EXTENDING 0.510 INCH ABOVE DATUM A, MAXIMUM MATERIAL CONDITION
[2.000] OR 2.000 BSC	THEORETICALLY EXACT DIMENSION IS 2.000

True Position Dimensioning Symbols
Figure 601

HINGE LINK PIN - REPAIR 1-1

66-14525–1, –2, –3, –4

1. General

A. This procedure has the data necessary to repair the hinge link pin.

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for the item numbers.

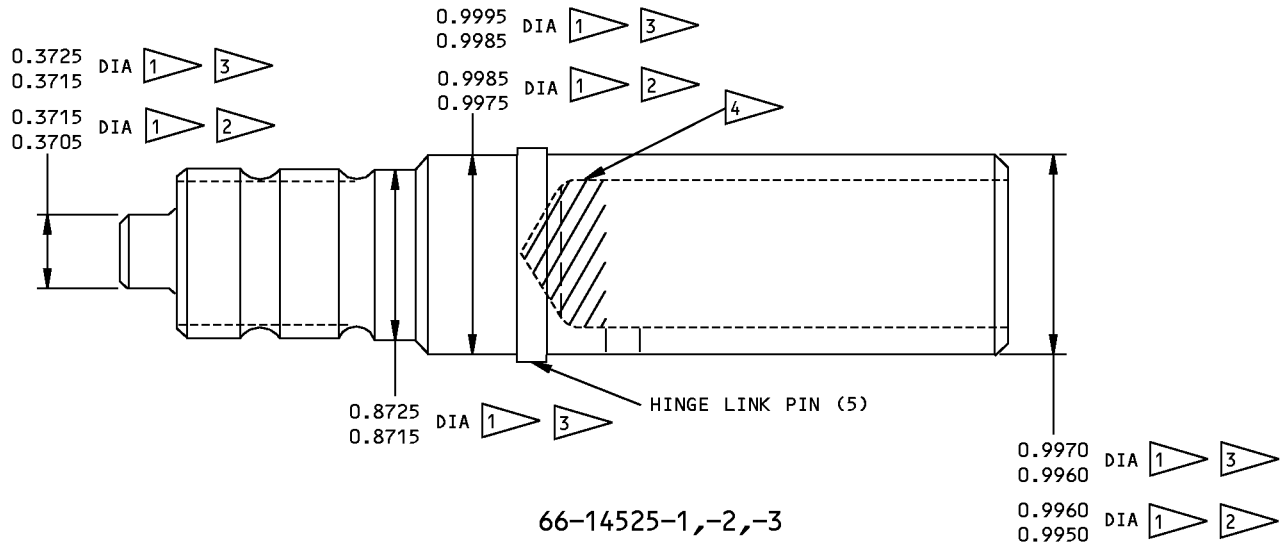
2. Refinish (REPAIR 1-1, Figure 601)

NOTE: Refer to REPAIR-GENERAL for a list of applicable standard procedures.

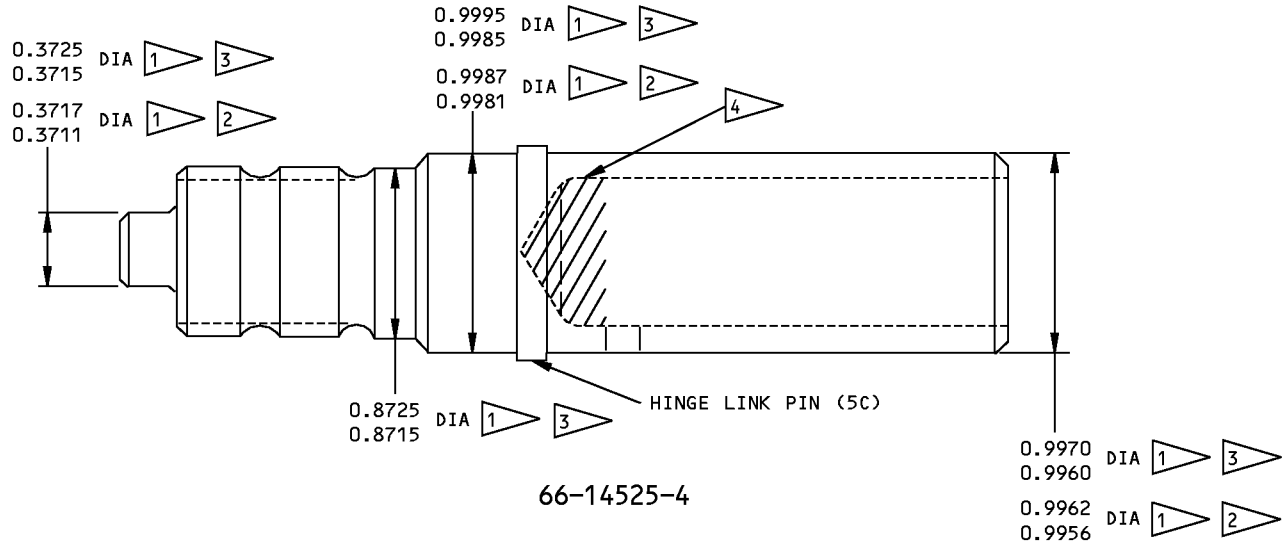
A. Pin (5) – Cadmium plate (F-1.913). On the inside diameter, apply primer, C00259 (SRF-12.206). Material: 4340 steel, 125-145 ksi.

B. Pin (5A) – Cadmium plate (F-16.04). On the inside diameter, apply phosphate coating (F-16.12, which replaces F-14.14) and primer, C00259 (F-20.03). Material: 4340 steel, 125-145 ksi.

C. Pin (5B, 5C) – Cadmium plate (F-15.02). Material: 15-5PH CRES, 150-170 ksi.



66-14525-1,-2,-3



66-14525-4

1 DIA TO BE CONCENTRIC WITHIN PLUS OR MINUS 0.001 T.I.R.

2 DIMENSION BEFORE PLATING

3 DIMENSION AFTER PLATING

4 FILL WITH LEVELING COMPOUND BMS 5-28, TYPE 1, EVEN WITH DRAIN HOLE AFTER FINISH IS APPLIED

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

276838 S00041002480_V3

66-14525-1,-2,-3,-4 Hinge Link Pin Refinish
Figure 601

HINGE LINK PIN - REPAIR 2-1

66-14526–1, –3, –5

1. General

A. This procedure has the data necessary to repair the hinge link pin.

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

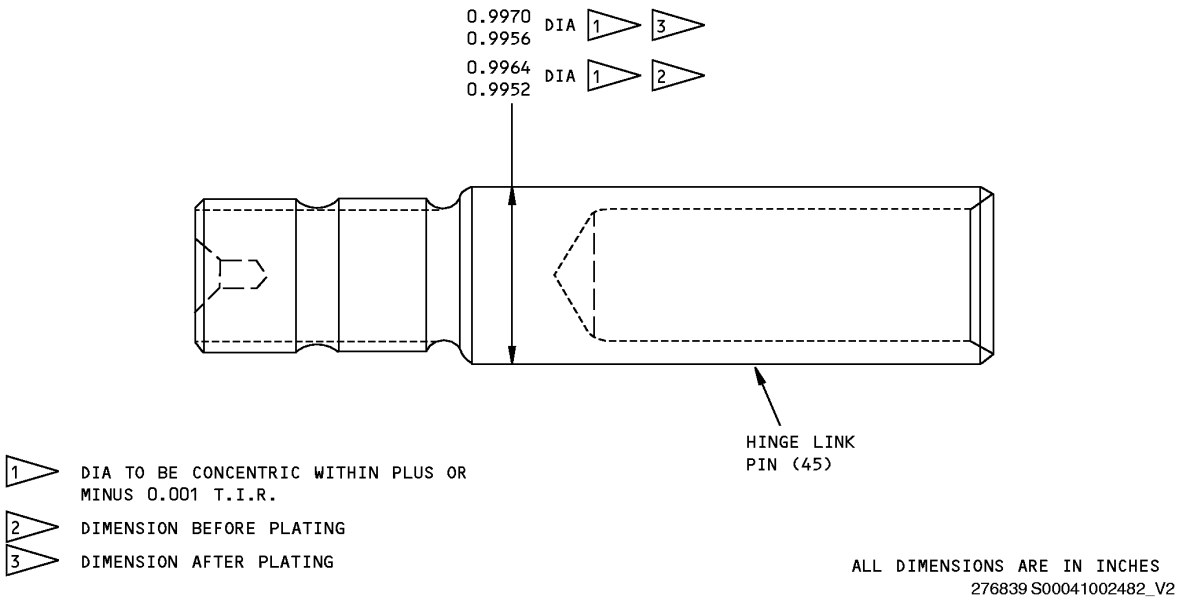
2. Refinish (REPAIR 2-1, Figure 601)

NOTE: Refer to REPAIR-GENERAL for a list of applicable standard procedures.

A. Pin (45) – Cadmium plate (F-1.913). On the inside diameter, apply primer, C00259 (SRF-12.206). Material: 4340 steel, 125-145 ksi.

B. Pin (45A) – Cadmium plate (F-16.04). On the inside diameter, apply phosphate coating (F-16.12, which replaces F-14.14) and primer, C00259 (F-20.03). Material: 4340 steel, 125-145 ksi.

C. Pin (45B) – Cadmium plate (F-15.06). Material: 15-5PH, 150-170 ksi.



66-14526-1,-3,-5 Hinge Link Pin Refinish
Figure 601

TORQUE TUBE - REPAIR 3-1

66-14537–2, –8, –13

1. General

A. This procedure has the data necessary to repair the torque tube.

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

2. Refinish

NOTE: Refer to REPAIR-GENERAL for a list of applicable standard procedures.

A. Tube (85)

(1) 4130 steel (150-170 ksi): Cadmium plate and apply primer, C00259 (SRF-1.611). After plating, design OD is 0.9956-0.9970 inch.

(2) 15-5PH CRES (150-170 ksi): Cadmium plate (F-15.05). After plating, design OD is 0.9956-0.9970 inch.

B. Tube (85A)

(1) 4130 steel (150-170 ksi): Cadmium plate and apply primer, C00259 (F-16.04). In ID, apply phosphate coating (F-16.12, which replaces F-14.14) and primer, C00259 (F-20.03). After plating, design OD is 0.9956-0.9970 inch.

(2) 15-5PH CRES (150-170 ksi): Cadmium plate (F-15.05). After plating, design OD is 0.9956-0.9970 inch.

C. Tube (85B) – Cadmium plate (F-15.02). Material: 17-7PH or 15-5PH CRES, 150-170 ksi.

TORQUE TUBE SLEEVE - REPAIR 4-1

66-23571–1, –2, –7, –8, –13, –14

1. General

A. This procedure has the data necessary to repair the torque tube.

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

2. Refinish

NOTE: Refer to REPAIR-GENERAL for a list of applicable standard procedures.

A. Sleeve (25, 65) – Exterior: Cadmium plate (F-1.32, which replaces F-1.1923). Material: 4340 steel, 150-170 ksi.

B. Sleeve (25A, 65A) – Exterior: Cadmium plate (F-16.04). Inside diameter: Apply phosphate coating (F-16.12, which replaces F-14.14) and primer, C00259 (0.0008 inch maximum) (F-20.03). Material: 4340 steel, 150-170 ksi.

C. Sleeve (25B, 65B) – Cadmium plate (F-15.06). Material: 17-7PH or 15-5PH CRES, 150-170 ksi.

ASSEMBLY

1. General

A. This procedure has the data necessary to assemble the forward galley door torque tube assembly.

B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for the item numbers.

2. Assembly

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
B50080	Compound - Corrosion Preventive, Solvent Cutback, Cold-Application (Grade 2 - Soft Film)	MIL-PRF-16173, Grade 2 (Supersedes MIL-C-16173, Grade 2)
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
D00633	Grease - Aircraft General Purpose	BMS3-33

B. References

Reference	Title
SOPM 20-50-01	BOLT AND NUT INSTALLATION
SOPM 20-60-02	FINISHING MATERIALS
SOPM 20-60-03	LUBRICANTS

C. Procedure

(1) Use standard industry practices and these steps.

NOTE: For bolt and nut installation, refer to SOPM 20-50-01. For finishing materials, refer to SOPM 20-60-02. For lubricants, refer to SOPM 20-60-03.

(a) Assemble to dimensions shown in ASSEMBLY, Figure 701.

(b) Apply a thin layer of corrosion preventive compound, B50080 to mating surfaces of sleeve (65) and torque tube (85). Install sleeve with bolt (70), washer (75), and nut (80). Tighten the nut (80) on the bolt (70) to a running torque of 30-40 pound-inches.

NOTE: For bolt (70A), the bolt direction is important to prevent interference and threads in bearing. The nut (80A) is intended to bottom out on the bolt threads to prevent clamp up. The maximum gap permitted between the sleeve (65) and the washers (75A) is 0.016 inch. If a large gap occurs, install more washers (75A) as necessary. Substitution of the bolt (70A) grip length is not permitted.

(c) Apply a thin layer of corrosion preventive compound, B50080 to mating surfaces of hinge link pin (45) and sleeve (65). Install hinge link pin with bolt (50), washer (55), and nut (60). Tighten the nut (60A) on the bolt (50A) to a running torque of 30-40 pound-inches.

NOTE: For bolt (50A), the bolt direction is important to prevent interferece and threads in bearing. The nut (60A) is intended to bottom out on the bolt threads to prevent clamp up. The maximum gap permitted between the sleeve (65) and the washers (55A) is 0.016 inch. If a larger gap occurs, install more washers (55A) as necessary. Substitution of the bolt (50A) grip length is not permitted.

(d) Apply a thin layer of corrosion preventive compound, B50080 to mating surfaces of sleeve (25) and torque tube (85). Install sleeve with bolt (30), washer (35), and nut (40). Tighten the nut (40A) on the bolt (30A) to a running torque of 30-40 pound-inches.

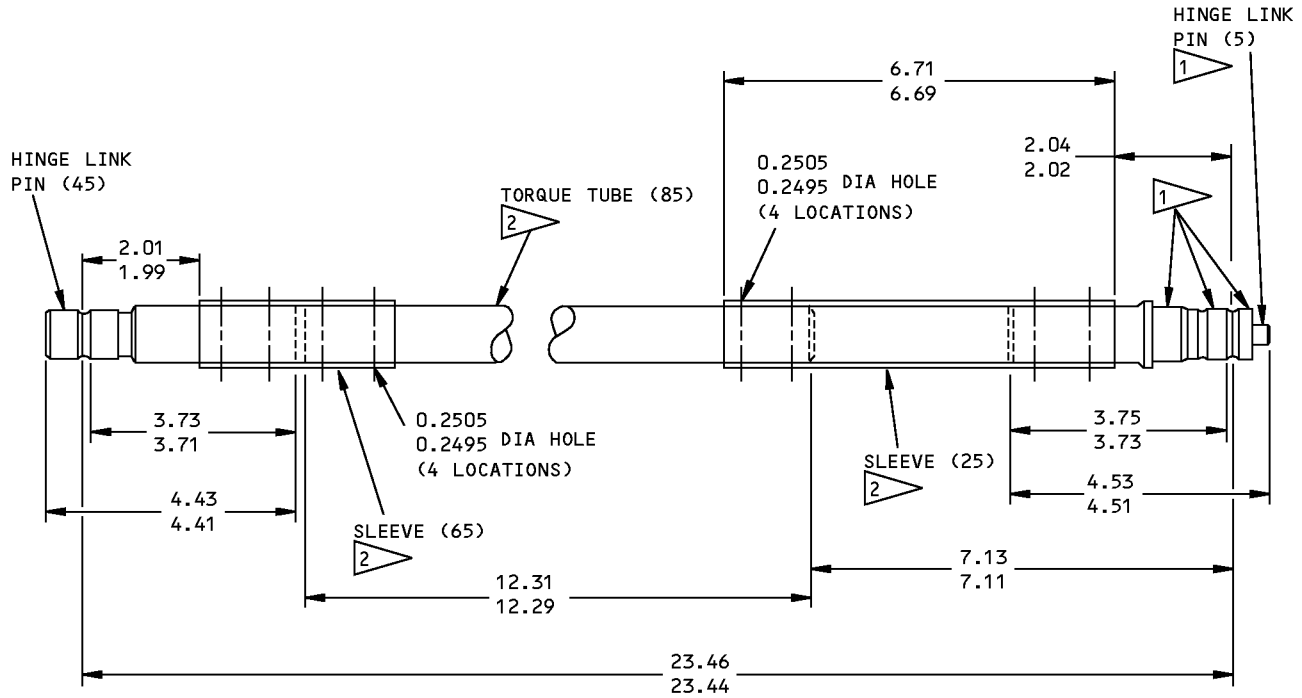
NOTE: For bolt (30A), the bolt direction is important to prevent interference and threads in bearing. The nut (40A) is intended to bottom out on the bolt threads to prevent clamp up. The maximum gap permitted between the sleeve (25) and the washers (35A) is 0.016 inch. If a larger gap occurs, install more washers (35A) as necessary. Substitution of the bolt (30A) grip length is not permitted.

(e) Apply a thin layer of corrosion preventive compound, B50080 to mating surfaces of hinge link pin (5) and sleeve (25). Install hinge link pin with bolt (10), washer (15), and nut (20). Tighten the nut (20A) on the bolt (10A) to a running torque of 30-40 pound-inches.

NOTE: For bolt (10A), the bolt direction is important to prevent interference and threads in bearing. The nut (20A) is intended to bottom out on the bolt threads to prevent clamp up. The maximum gap permitted between the sleeve (25) and the washers (15A) is 0.016 inch. If a larger gap occurs, install more washers (15A) as necessary. Substitution of the bolt (10A) grip length is not permitted.

(f) Apply a thin layer of corrosion preventive compound, B50080 to surfaces of mating threads on pins (5, 45).

(g) Apply a thin layer of grease, D00633 or grease, D00013 to the surfaces shown in ASSEMBLY, Figure 701.



1 APPLY A THIN LAYER OF GREASE

2 ASSEMBLE WITH A THIN LAYER OF MIL-C-16173 GRADE 2 CORROSION PREVENTIVE COMPOUND

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

275255 S00041002486_V3

Torque Tube Assembly
Figure 701

BOEING®

COMPONENT MAINTENANCE MANUAL

FITS AND CLEARANCES

(NOT APPLICABLE)

(NOT APPLICABLE)

ILLUSTRATED PARTS LIST

1. Introduction

A. The Illustrated Parts List (IPL) contains an illustration and a list of component parts you can repair or replace. The Illustrated Parts Catalog (IPC) shows how to use the Boeing part number system.

B. This shows how parts are related: The relation of each item to its next higher assembly (NHA) is shown in the NOMENCLATURE column. Use the indenture system that follows:

1 2 3 4 5 6 7

. Assembly

. Attaching parts for assembly

. . Detail parts for assembly

. . Subassembly

. . Attaching parts for subassembly

. . . Detail parts for subassembly

. . . Sub-subassembly

. . . Attaching parts for subassembly

. . . . . Details parts for sub-subassembly

Detail Installation Parts (Included only if installation parts may be sent to the shop as part of assembly)

C. Each top assembly is given one use code letter (A, B, C, etc.) in the USAGE CODE column. All subsequent component parts in the list can have one or more of the use code letters to show effectivity to top assemblies. A component part without a use code applies to all top assemblies.

D. An alphabetical letter is added after the item number for optional parts, parts changed by a Service Bulletin, configuration differences (except left-handed and right-handed parts), last engineering releases, and parts added between item numbers in a sequence. The alphabetical letter will not be shown on the illustration for equivalent parts of the same part number.

E. Color-coded parts are identified with a single digit alpha following the dash number or with ''SP'' suffix. If the ''SP'' suffix is used, it represents consolidation of all color codes applicable for a given usage which are not separately listed. Orders for color-coded parts should include the registry number of the airplane for which the parts are ordered.

F. If a part number is 15 characters long but will not fit in the part number column, the part number will be displayed with a ''~'' at the end of the line and will be continued on the next line. The ''~'' denotes that the part number continues on the next line.

G. Parts changed by a Service Bulletin are shown by PRE SB XXXX and POST SB XXXX added to the NOMENCLATURE column.

(1) When a new top assembly is added by a Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the top assembly level only. The configuration differences at the detail part level are shown by use code letters.

(2) When the top assembly part number is not changed by the Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the detail level.

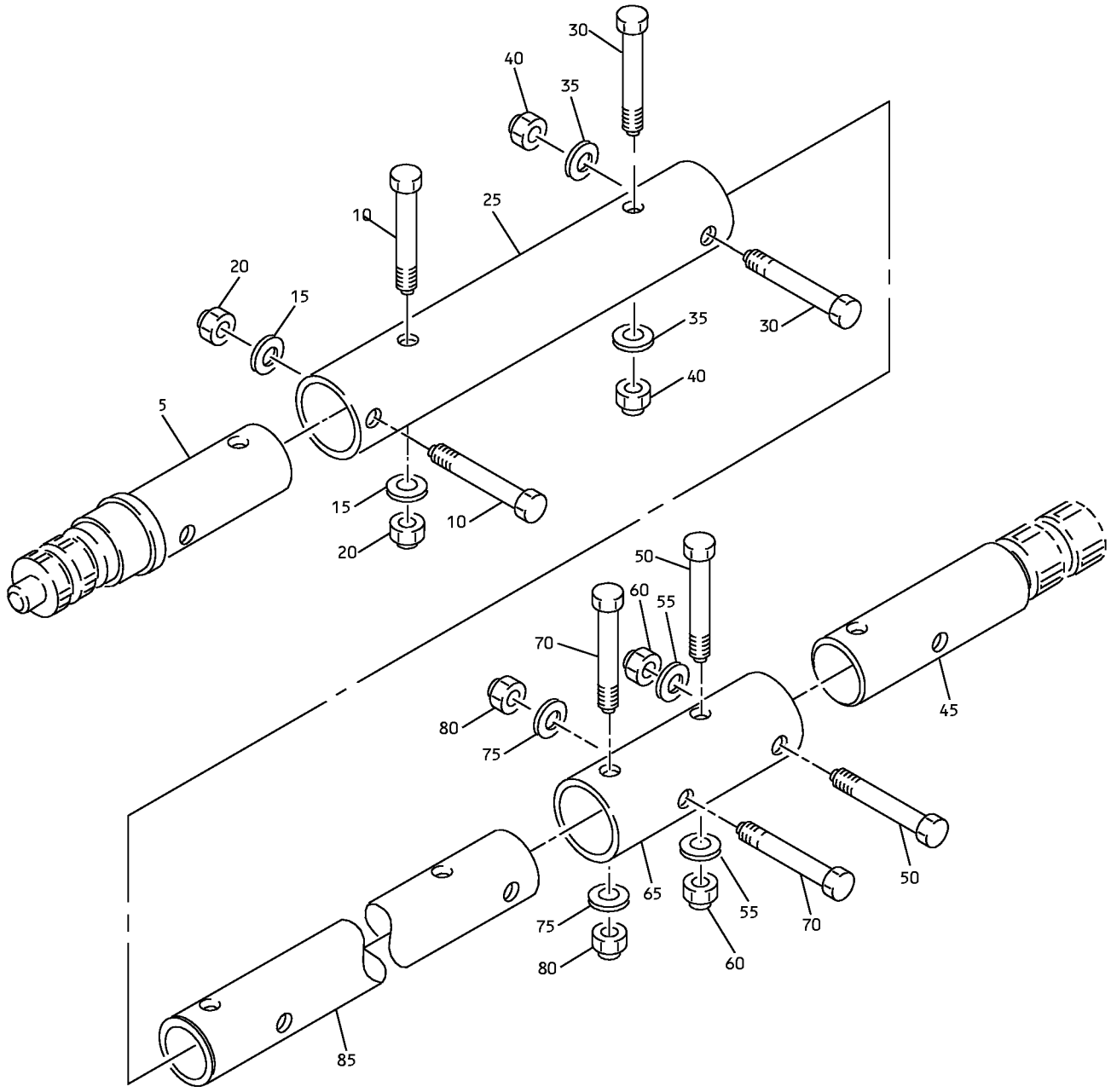
H. Interchangeable Parts

Optional (OPT)	The part is optional to and interchangeable with other parts that have the same item number.
Replaces, Replaced by and not interchangeable with (REPLACES, REPLACED BY AND NOT INTCHG/W)	The part replaces and is not interchangeable with the initial part.
Replaces, Replaced by (REPLACES, REPLACED BY)	The part replaces and is interchangeable with, or is an alternative to, the initial part.

NUMERICAL INDEX

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
65-45871-102		1	1B	RF
65-45871-113		1	1C	RF
65-45871-142		1	1D	RF
65-45871-88		1	1A	RF
66-14525-1		1	5	1
66-14525-2		1	5A	1
66-14525-3		1	5B	1
66-14525-4		1	5C	1
66-14526-1		1	45	1
66-14526-3		1	45A	1
66-14526-5		1	45B	1
66-14537-13		1	85B	1
66-14537-2		1	85	1
66-14537-8		1	85A	1
66-23571-1		1	25	1
66-23571-13		1	25B	1
66-23571-14		1	65B	1
66-23571-2		1	65	1
66-23571-7		1	25A	1
66-23571-8		1	65A	1
AN960C416L		1	15A	2
		1	35A	2
		1	55A	2
		1	75A	2
AN960PD416L		1	15	2
		1	35	2
		1	55	2
		1	75	2
BACB30NM4K20		1	10A	2
		1	30A	2
		1	50A	2
BACB30NMM4K20		1	70A	2
BACN10YR4CD		1	20A	2
		1	40A	2

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		1	60A	2
		1	80A	2
NAS1104-19		1	10	2
		1	30	2
		1	50	2
		1	70	2
NAS679A4W		1	20	2
		1	40	2
		1	60	2
		1	80	2



275499 S00041002489_V3

Forward Galley Door Torque Tube Assembly
IPL Figure 1

FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USAGE CODE	UNITS PER ASSY
1–					
–1A	65-45871-88		TORQUE TUBE ASSY-FWD GALLEY DOOR (PRE SB 52-1094)	A	RF
–1B	65-45871-102		TORQUE TUBE ASSY-FWD GALLEY DOOR (PRE SB 52-1094)	B	RF
–1C	65-45871-113		TORQUE TUBE ASSY-FWD GALLEY DOOR (POST SB 52-1094)	C	RF
–1D	65-45871-142		TORQUE TUBE ASSY-FWD GALLEY DOOR	D	RF
5	66-14525-1		. PIN, HINGE LINK	A	1
–5A	66-14525-2		. PIN, HINGE LINK	B	1
–5B	66-14525-3		. PIN, HINGE LINK	C	1
–5C	66-14525-4		. PIN, HINGE LINK	D	1
10	NAS1104-19		. BOLT	A, B	2
–10A	BACB30NM4K20		. BOLT	C, D	2
15	AN960PD416L		. WASHER	A, B	2
–15A	AN960C416L		. WASHER	C, D	2
20	NAS679A4W		. NUT	A, B	2
–20A	BACN10YR4CD		. NUT	C, D	2
25	66-23571-1		. SLEEVE	A	1
–25A	66-23571-7		. SLEEVE	B	1
–25B	66-23571-13		. SLEEVE	C, D	1
30	NAS1104-19		. BOLT	A, B	2
–30A	BACB30NM4K20		. BOLT	C, D	2
35	AN960PD416L		. WASHER	A, B	2
–35A	AN960C416L		. WASHER	C, D	2
40	NAS679A4W		. NUT	A, B	2
–40A	BACN10YR4CD		. NUT	C, D	2
45	66-14526-1		. PIN, HINGE LINK	A	1
–45A	66-14526-3		. PIN, HINGE LINK	B	1
–45B	66-14526-5		. PIN, HINGE LINK	C, D	1

–Item not Illustrated

FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USAGE CODE	UNITS PER ASSY
1–					
50	NAS1104-19		. BOLT	A, B	2
–50A	BACB30NM4K20		. BOLT	C, D	2
55	AN960PD416L		. WASHER	A, B	2
–55A	AN960C416L		. WASHER	C, D	2
60	NAS679A4W		. NUT	A, B	2
–60A	BACN10YR4CD		. NUT	C, D	2
65	66-23571-2		. SLEEVE	A	1
–65A	66-23571-8		. SLEEVE	B	1
–65B	66-23571-14		. SLEEVE	C, D	1
70	NAS1104-19		. BOLT	A, B	2
–70A	BACB30NMM4K20		. BOLT	C, D	2
75	AN960PD416L		. WASHER	A, B	2
–75A	AN960C416L		. WASHER	C, D	2
80	NAS679A4W		. NUT	A, B	2
–80A	BACN10YR4CD		. NUT	C, D	2
85	66-14537-2		. TUBE, TORQUE	A	1
–85A	66-14537-8		. TUBE, TORQUE	B	1
–85B	66-14537-13		. TUBE, TORQUE	C, D	1

–Item not Illustrated