COMPONENT MAINTENANCE MANUAL WITH ILLUSTRATED PARTS LIST

FORWARD ACCESS DOOR ASSEMBLY

PART NUMBER 141A6802–3, –4, –5, –6

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COMPONENT MAINTENANCE MANUAL

Revision No. 9
Jul 01/2009

To: All holders of FORWARD ACCESS DOOR ASSEMBLY 52-48-01.

Attached is the current revision to this COMPONENT MAINTENANCE MANUAL

The COMPONENT MAINTENANCE MANUAL is furnished either as a printed manual, on microfilm, or digital products, or any combination of the three. This revision replaces all previous microfilm cartridges or digital products. All microfilm and digital products are reissued with all obsolete data deleted and all updated pages added.

For printed manuals, changes are indicated on the List of Effective Pages (LEP). The pages which are revised will be identified on the LEP by an R (Revised), A (Added), O (Overflow, i.e. changes to the document structure and/or page layout), or D (Deleted). Each page in the LEP is identified by Chapter-Section-Subject number, page number and page date.

Pages replaced or made obsolete by this revision should be removed and destroyed.

ATTENTION

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Location of Change	Description of Change
52-48-01	
REPAIR 4-1	Changed the data in the Consumable Materials list.
ASSEMBLY	Changed the data in the Consumable Materials list.

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A = Added, R = Revised, D = Deleted, O = Overflow

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COMPONENT MAINTENANCE MANUAL

TEMPORARY REVISION AND SERVICE BULLETIN RECORD

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVE	DATE OF INCORPORATION INTO MANUAL

All revisions to this manual will be accompanied by transmittal sheet bearing the revision number. Enter the revision number in numerical order, together with the revision date, the date filed and the initials of the person filing.

Revision		Filed		Revision		Filed	
Number	Date	Date	Initials	Number	Date	Date	Initials

Revision		Filed		Revision		Filed	
Number	Date	Date	Initials	Number	Date	Date	Initials

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All temporary revisions to this manual will be accompanied by a cover sheet bearing the temporary revision number. Enter the temporary revision number in numerical order, together with the temporary revision date, the date the temporary revision is inserted and the initials of the person filing.
When the temporary revision is incorporated or cancelled, and the pages are removed, enter the date the pages are removed and the initials of the person who removed the temporary revision.

Temporary Revision		Inserted		Removed	
Number	Date	Date	Initials	Date	Initials

Temporary Revision		Inserted		Removed	
Date	Initials	Number	Date	Date	Initials

Temporary Revision		Inserted		Removed	
Number	Date	Date	Initials	Date	Initials

Temporary Revision		Inserted		Removed	
Date	Initials	Number	Date	Date	Initials

INTRODUCTION

1. General

A. The instructions in this manual supply the data necessary to do the maintenance functions together with the test, fault isolation, repair, and replacement of the defective parts.

B. This manual is divided into different parts:

(1) Title Page

(2) Transmittal Letter

(3) Highlights

(4) List of Effective Pages

(5) Table of Contents

(6) Temporary Revision & Service Bulletin Record

(7) Record of Revisions

(8) Record of Temporary Revisions

(9) Introduction

(10) Procedures & IPL Sections

C. Components that can be repaired have a different repair number for each specified repair. To find the repair number location of a component, look in the Repair-General procedure at the beginning of the REPAIR section. The Repair-General procedure also has an explanation of the True Position Dimension symbols used.

D. All dimensions, measures, quantities and weights included are in English units. When metric equivalents are given they will be in the parentheses that follow the English units.

E. The introduction to the Illustrated Parts List (IPL) shows how the IPL data is used.

F. Design changes, optional parts, configuration differences and Service Bulletin modifications may cause different part numbers. These part numbers are identified in the IPL with an alphabetical letter which is added to the end of the basic item number. This new item number is referred to as an alpha-variant. Throughout the manual, IPL basic item number references also apply to alpha-variants unless shown differently.

G. The tool reference numbers found in the individual procedures and in the Special Tools, Fixtures, and Equipment section are used to identify if a tool is a standard tool (STD-XXXX), a commercial tool (COM-XXXX), or a Special Tool (SPL-XXXX). This reference number is also used to distinguish between tools with similar names in the same procedure. These reference numbers are for use in the documentation only. They are not to be used for ordering tools.

FORWARD ACCESS DOOR ASSEMBLY - DESCRIPTION AND OPERATION

1. Description

A. The forward access door (DESCRIPTION AND OPERATION, Figure 1) permits access to the forward equipment bay for maintenance and the repair of equipment in the forward equipment bay. The forward access door is attached to the fuselage by two hinges on the aft edge and a latch pin on the forward edge.

2. Operation

A. The handle assembly is found in the center of the door. The handle assembly releases the latch pin and permits a person to open the forward access door.

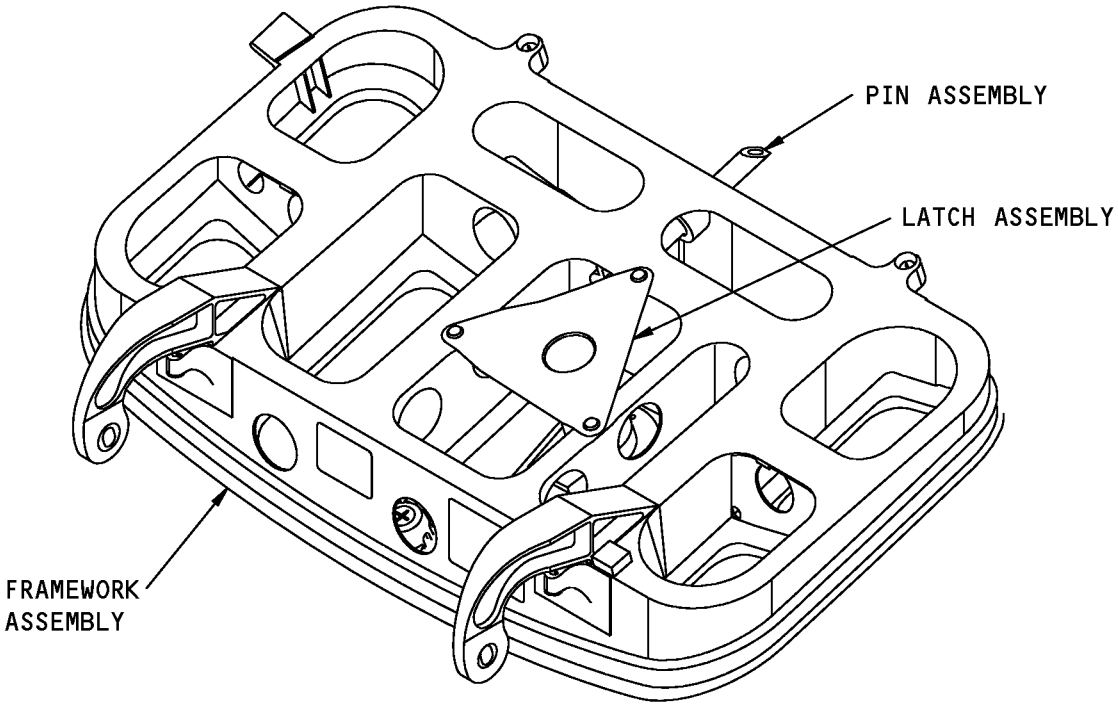
3. Leading Particulars (Approximate)

A. Length – 31.0 inches

B. Width – 22.0 inches

C. Height – 7.0 inches

D. Weight – 13.5 pounds



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Forward Access Door Assembly
Figure 1

(NOT APPLICABLE)

DISASSEMBLY

1. **General**
   A. This procedure has the data necessary to disassemble the forward access door assembly.
   B. Disassemble this component sufficiently to isolate the defects, do the necessary repairs, and put the component back to a serviceable condition.
   C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
2. **Disassembly**
   A. Procedure
      (1) Use standard industry procedures to disassemble this component.

CLEANING

1. **General**
   A. This procedure has the data necessary to clean the forward access door assembly.
   B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
   C. Refer to IPL Figure 1 for item numbers.
2. **Cleaning**
   A. Procedure
      (1) Clean all the parts other than the bearing (135) as specified by standard industry practices (SOPM 20-30-03).
      (2) Clean the teflon-sealed bearing (135) as specified by the manufacturer's instructions.

CHECK

1. **General**
   A. This procedure has the data necessary to find defects in the material of the specified parts.
   B. Refer to FITS AND CLEARANCES for the design dimension and wear limits.
   C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
   D. Refer to IPL Figure 1 for item numbers.
2. **Check**
   A. Procedure
      (1) Use standard industry procedures to do a visual check of all the parts for defects. Do the penetrant or magnetic particle check if the visual check shows possible damage or if you suspect possible damage on the parts listed below:
      (2) Do a magnetic particle check (SOPM 20-20-01) of these parts:
         (a) Latch pin (140)
      (3) Do a penetrant check (SOPM 20-20-02) of these parts:
         (a) Framework (190)

REPAIR

1. General

A. Instructions for repair, refinish, and replacement of the specified subassembly parts are included in each REPAIR when applicable:

Table 601:

PART NUMBER	NAME	REPAIR
—	REFINISH OF OTHER PARTS	1-1
141A6802	FORWARD ACCESS DOOR ASSEMBLY	2-1
141A6803 141A6815	FRAMEWORK ASSEMBLY	3-1
141A6804 141A6816	FRAMEWORK	3-2
141A6810	LATCH PIN ASSEMBLY	4-1

2. Dimensioning Symbols

A. Standard True Position Dimensioning Symbols used in the applicable repair procedures are shown in REPAIR-GENERAL, Figure 601.

—	STRAIGHTNESS
▱	FLATNESS
⊥	PERPENDICULARITY (OR SQUARENESS)
//	PARALLELISM
○	ROUNDNESS
⌭	CYLINDRICITY
⌒	PROFILE OF A LINE
⌓	PROFILE OF A SURFACE
◎	CONCENTRICITY
⌯	SYMMETRY
∠	ANGULARITY
↗	RUNOUT
↗↗	TOTAL RUNOUT
⌴	COUNTERBORE OR SPOTFACE
∨	COUNTERSINK
⌖	THEORETICAL EXACT POSITION OF A FEATURE (TRUE POSITION)
∅	DIAMETER
S ∅	SPHERICAL DIAMETER
R	RADIUS
SR	SPHERICAL RADIUS
()	REFERENCE
BASIC (BSC) OR [DIM]	A THEORETICALLY EXACT DIMENSION USED TO DESCRIBE SIZE, SHAPE OR LOCATION OF A FEATURE. FROM THIS FEATURE PERMISSIBLE VARIATIONS ARE ESTABLISHED BY TOLERANCES ON OTHER DIMENSIONS OR NOTES.
[-A-]	DATUM
Ⓜ	MAXIMUM MATERIAL CONDITION (MMC)
Ⓛ	LEAST MATERIAL CONDITION (LMC)
Ⓢ	REGARDLESS OF FEATURE SIZE (RFS)
Ⓟ	PROJECTED TOLERANCE ZONE
FIM	FULL INDICATOR MOVEMENT

EXAMPLES

— | 0.002	STRAIGHT WITHIN 0.002
⊥ | 0.002 | B	PERPENDICULAR TO DATUM B WITHIN 0.002
// | 0.002 | A	PARALLEL TO DATUM A WITHIN 0.002
○ | 0.002	ROUND WITHIN 0.002
⌭ | 0.010	CYLINDRICAL SURFACE MUST LIE BETWEEN TWO CONCENTRIC CYLINDERS, ONE OF WHICH HAS A RADIUS 0.010 INCH GREATER THAN THE OTHER
⌒ | 0.006 | A	EACH LINE ELEMENT OF THE SURFACE AT ANY CROSS SECTION MUST LIE BETWEEN TWO PROFILE BOUNDARIES 0.006 INCH APART RELATIVE TO DATUM A
⌓ | 0.020 | A	SURFACES MUST LIE WITHIN PARALLEL BOUNDARIES 0.020 INCH APART AND EQUALLY DISPOSED ABOUT TRUE PROFILE
◎ | ∅ 0.0005 | C	CONCENTRIC TO DATUM C WITHIN 0.0005 DIAMETER
⌯ | 0.010 | A	SYMMETRICAL WITH DATUM A WITHIN 0.010
∠ | 0.005 | A	ANGULAR TOLERANCE 0.005 WITH DATUM A
⌖ | ∅ 0.002 Ⓢ | B	LOCATED AT TRUE POSITION WITHIN 0.002 DIA RELATIVE TO DATUM B, REGARDLESS OF FEATURE SIZE
⊥ | ∅ 0.010 Ⓜ | A 0.510 Ⓟ	AXIS IS TOTALLY WITHIN A CYLINDER OF 0.010 INCH DIAMETER, PERPENDICULAR TO DATUM A, AND EXTENDING 0.510 INCH ABOVE DATUM A, MAXIMUM MATERIAL CONDITION
[2.000] OR 2.000 BSC	THEORETICALLY EXACT DIMENSION IS 2.000

True Position Dimensioning Symbols
Figure 601

REFINISH OF OTHER PARTS - REPAIR 1-1

1. General

A. This procedure has the data necessary to refinish the parts which are not given in the specified repairs.

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

2. Refinish of Other Parts

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

C. General

(1) Instructions for the repair of the parts listed in REPAIR 1-1, Table 601 is for repair of the initial finish.

D. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For the decoding table for Boeing finish codes, refer to SOPM 20-41-01. For finishing materials, refer to SOPM 20-60-02.

(1) Refer to REPAIR 1-1, Table 601 for refinish details.

Table 601: Refinish Details

IPL FIG. & ITEM	MATERIAL	FINISH
IPL Fig. 1		
Plate, Handle support (30)	Aluminum alloy	Boric acid-sulfuric acid andoize or chromic acid anodize (F-17.31) and apply primer, C00259 (F-20.03).
Linkage, Latch (115)	Aluminum alloy	Boric acid-sulfuric acid anodize or chromic acid anodize (F-17.31) and apply primer, C00259 (F-20.03) except do not apply primer into holes.

FORWARD ACCESS DOOR ASSEMBLY - REPAIR 2-1

141A6802–3, –4, –5, –6

1. General

A. This procedure has the data necessary to repair and refinish the forward access door assembly (1A, 1B, 1C, 1D).

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.

D. Refer to IPL Figure 1 for item numbers.

2. Bumper Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00153	Adhesive - Low Odor, Synthetic Rubber, 1 Part	BMS 5-30

B. References

Reference	Title
SOPM 20-30-03	GENERAL CLEANING PROCEDURES
SOPM 20-50-12	APPLICATION OF ADHESIVES

C. Procedure

NOTE: Equivalent material can be used.

(1) Remove the bumper (35) from the frame work assembly (155).

(2) Clean bonding surface as shown in SOPM 20-30-03.

(3) Apply adhesive, A00153 onto the bonding surface of the frame work assembly (155) as shown in SOPM 20-50-12.

(4) Install the new bumper (35) onto the frame work assembly (155).

3. Seal Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
B00003	Cleaner - Emulsion Alkaline - GMC 528B	
B00052	Soap - Liquid - Turco 1526	BAC5507
B50093	Soap - Liquid - Kelite Spraywhite	BAC5507

B. References

Reference	Title
BAC 5507	Installation of Air Seals
SOPM 20-60-01	CLEANING MATERIALS

C. Procedure

NOTE: For cleaning materials, refer to SOPM 20-60-01.

(1) Remove the seal (40B) from the frame work assembly (155).

NOTE: Use Kelite Spraywhite, B50093, Turco 1526 soap, B00052 or GMC 528B cleaner, B00003 (SOPM 20-60-01) to install the seal (40B), if necessary.

(2) Install the new seal (40B) onto the frame work assembly (155). Refer to ASSEMBLY and BAC 5507 for the seal (40B) installation procedures.

4. Forward Access Door Assembly Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I
C50069	Coating - Enamel, Color 702 Gloss White	BMS10-11, Type II

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-41-02	APPLICATION OF CHEMICAL AND SOLVENT RESISTANT FINISHES
SOPM 20-43-03	CHEMICAL CONVERSION COATINGS FOR ALUMINUM
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure (REPAIR 2-1, Figure 601)

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For the decoding table for Boeing finish codes, refer to SOPM 20-41-01 For finishing materials, refer to SOPM 20-60-02.

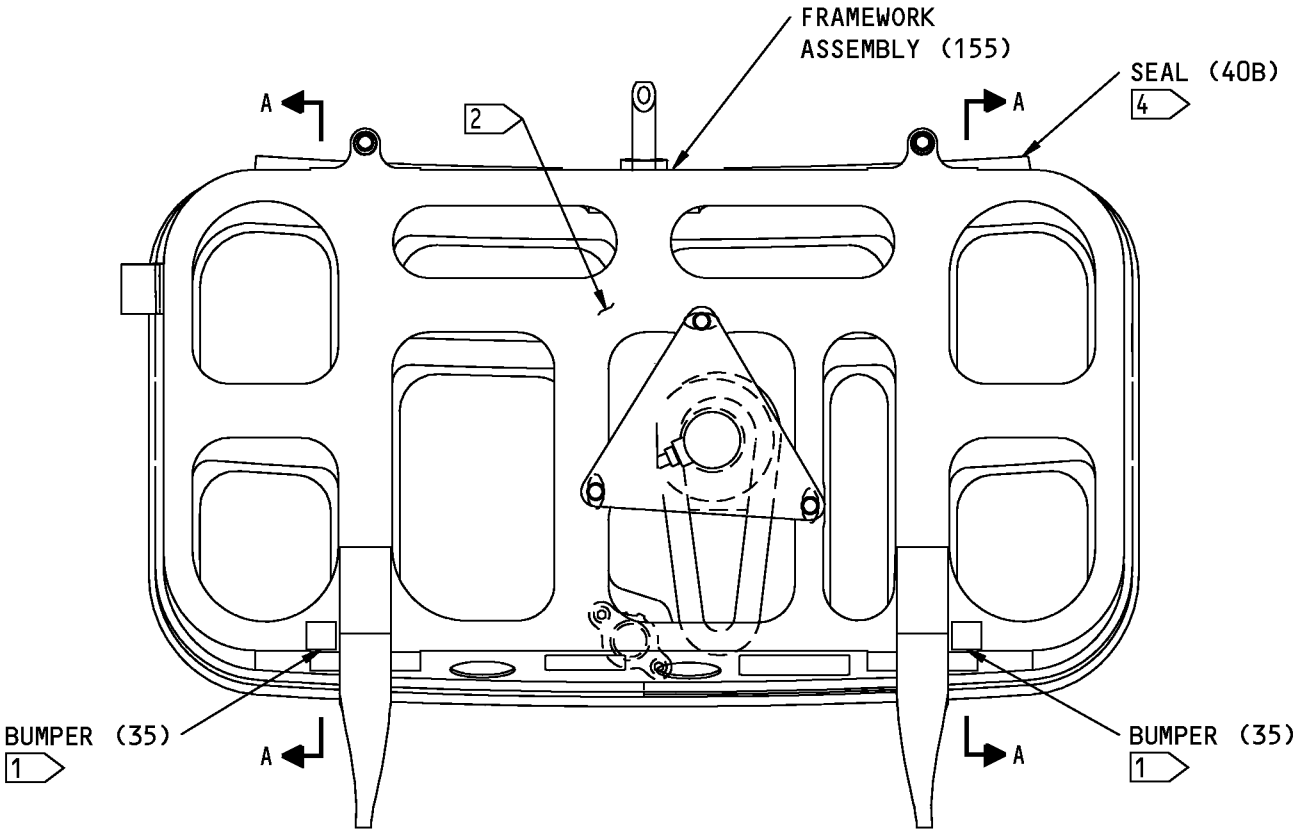
(1) Chemical treat scratches and bare surfaces (F-14.995) as shown in SOPM 20-43-03.

(2) Clean and reactivate primed surfaces as shown in SOPM 20-41-02.

(3) Apply yellow primer, C00259 (F-14.995), as shown in SOPM 20-41-02. Do not apply the primer to the outside surface of the outer skin, seal, stop pins, grounding studs, handle and latch mechanism.

(4) Apply enamel coating, C50069 (F-21.03) per REPAIR 2-1, Figure 601, flagnote 3 and SOPM 20-41-02.

(5) Obey all flagnotes.



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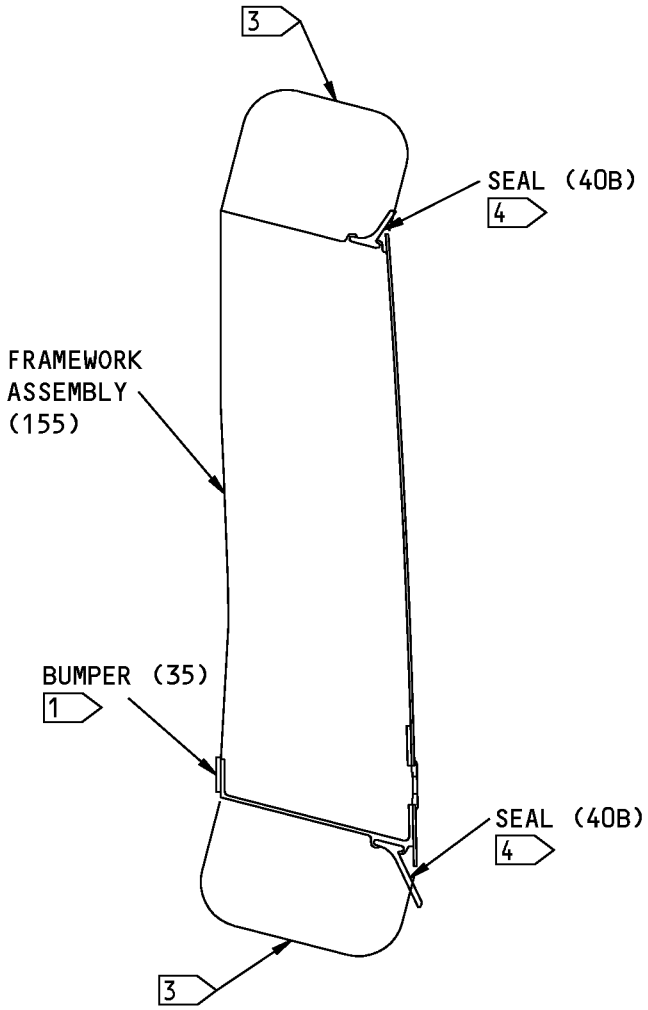
141A6802-3, -4, -5, -6 Forward Access Door Assembly Repair
Figure 601 (Sheet 1 of 2)



1 INSTALL BUMPER WITH BMS 5-30, TYPE 48 ADHESIVE AS SHOWN IN SOPM 20-50-12

2 APPLY PANEL IDENTIFICAION NUMBER -112A- TO THE INTERNAL SURFACE OF THIS PANEL AFTER APPLICATION OF ALL FINISHES. APPLICATION IS CONTROLLED BY DRAWING NUMBER 141A0042

3 APPLY BMS 10-11, TYPE 1 PRIMER (F-14.9962) AND BMS 10-11, TYPE 2 ENAMEL (F-21.03) ALL AROUND THE OUTER FRAME. INCLUDE STOPS AND BOTH SIDES OF THE HINGES BEFORE INSTALLATION OF SEAL. DO NOT APPLY ANY FINISH TO STOP AND HINGE BUSHINGS. FULL COVERAGE NOT REQUIRED INSIDE SEAL RETAINER GROOVE. OVERSPRAY IS NOT ALLOWED ON OUTSIDE SURFACE OF THE SKIN. OVERSPRAY PERMISSIBLE ON INNER SURFACES OF THE DOOR

4 SEAL MAY BE INSTALLED WITH SOAP LUBRICANTS AS SHOWN IN SOPM 20-60-01

ITEM NUMBER REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

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141A6802-3, -4, -5, -6 Forward Access Door Assembly Repair
Figure 601 (Sheet 2 of 2)

FRAMEWORK ASSEMBLY - REPAIR 3-1

141A6803–2, –3, 141A6815–1

1. General

A. This procedure has the data necessary to repair and refinish the framework assembly (155).

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.

D. Refer to IPL Figure 1 for item numbers.

2. Stop Plate Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

B. References

Reference	Title
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure

NOTE: For miscellaneous materials, refer to SOPM 20-60-04.

(1) Remove the cotter pin (180) and the plate stop (185) from the framework (190).

(2) Install the plate stop (185) with sealant, A00247 as shown in SOPM 20-60-04 onto the framework (190).

(3) Install the cotter pin (180) onto the plate stop (185).

3. Bushing Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

B. References

Reference	Title
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure (REPAIR 3-1, Figure 601)

NOTE: For miscellaneous materials, refer to SOPM 20-60-04.

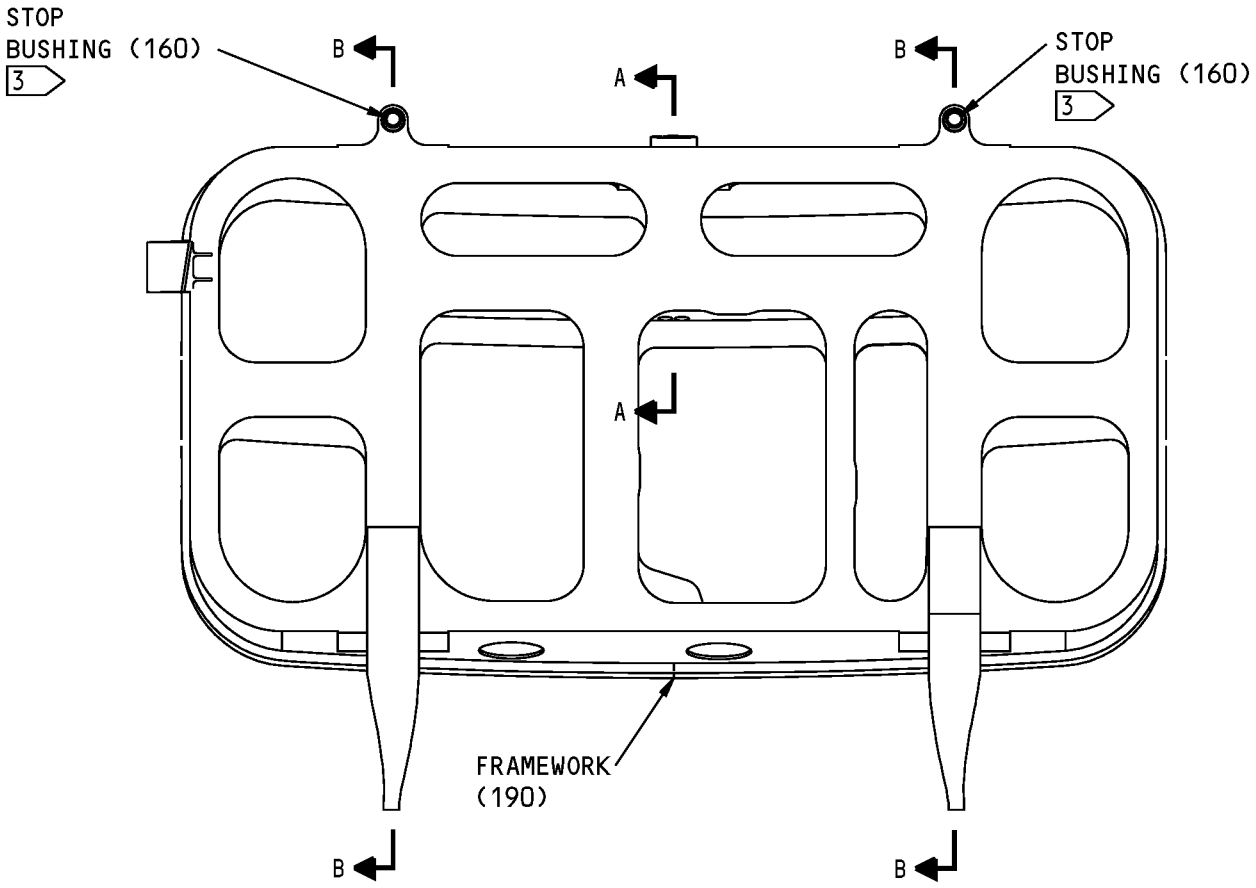
(1) Remove the bushings (160, 165, 170, 175) from the framework (190).

(2) Install the bushing (160) with wet sealant, A00247 as shown in SOPM 20-50-03. Flare the bushing (160) to align with the 30 degree countersink in the framework (190) before the sealant, A00247 dries.

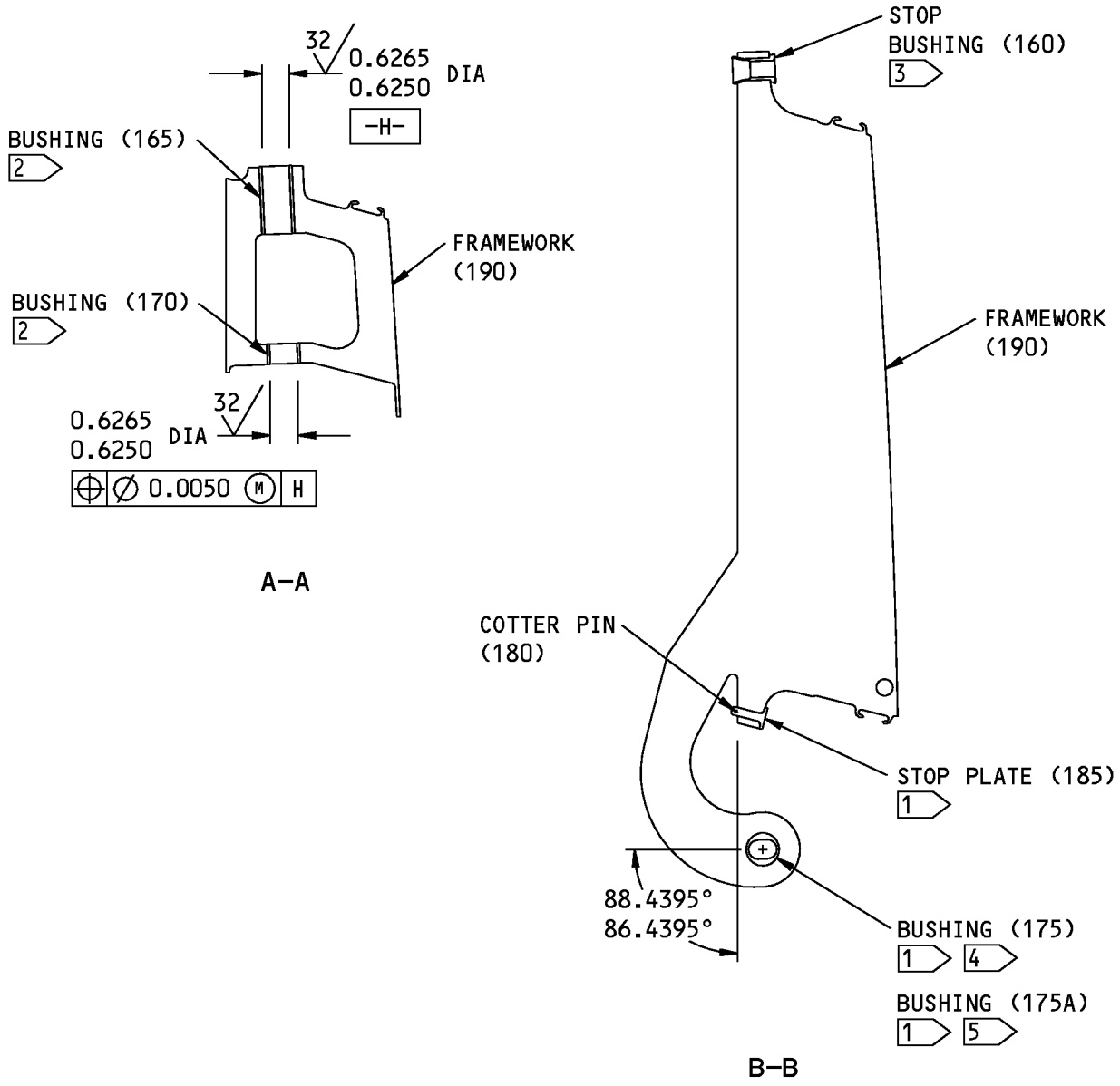
NOTE: The flared bushing (160) must touch the framework (190) with a tight fit.

(3) Install the bushings (165, 170, 175) by shrink-fit procedure with wet sealant, A00247 as shown in SOPM 20-50-03.

(4) Obey all flagnotes.



141A6803-2, -3, 141A6815-1 Framework Assembly Repair
Figure 601 (Sheet 1 of 2)



1 INSTALL WITH WET BMS 5-95 SEALANT

2 INSTALL BY SHRINK FIT METHOD WITH WET BMS 5-95 SEALANT

3 INSTALL WITH WET BMS 5-95 SEALANT AND FLARE

4 141A6803-2

5 141A6803-3,141A6815-1

ITEM NUMBER REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

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141A6803-2, -3, 141A6815-1 Framework Assembly Repair
Figure 601 (Sheet 2 of 2)

FRAMEWORK - REPAIR 3-2

141A6804–2, 141A6816–1

1. General

A. This procedure has the data necessary to repair and refinish the framework (190).

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.

D. Refer to IPL Figure 1 for item numbers.

2. Framework Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I

B. References

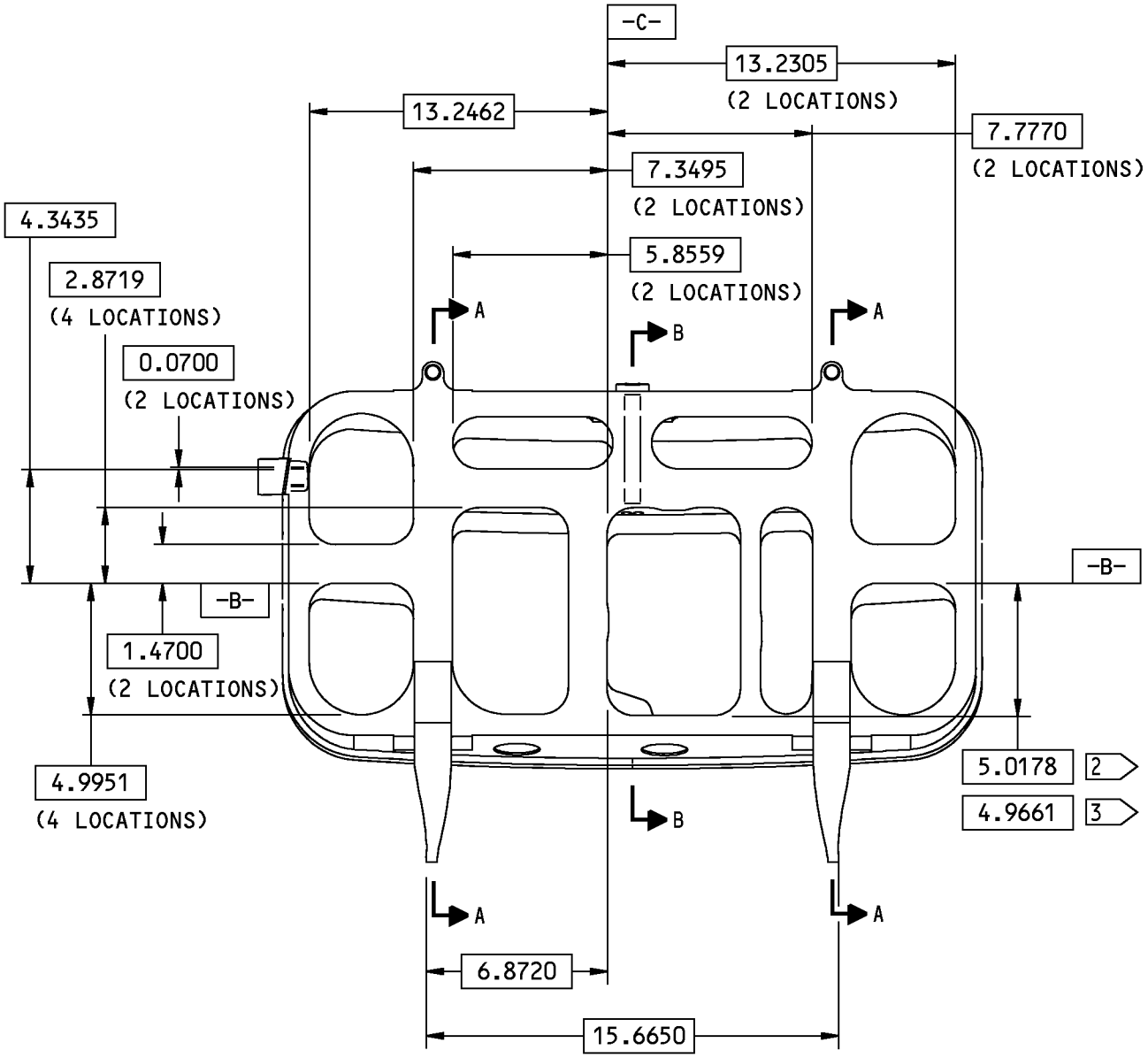
Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-43-01	CHROMIC ACID ANODIZING
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure (REPAIR 3-2, Figure 601)

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For the decoding table for Boeing finish codes, refer to SOPM 20-41-01. For chromic acid anodizing, refer to SOPM 20-43-01. For finishing materials, refer to SOPM 20-60-02.

(1) Chromic acid anodize or boric acid-sulfuric acid anodize (F-17.31) and then seal in chromate solution.

(2) Apply primer, C00259 (F-20.03). Do not apply the primer, C00259 to holes identified by flagnote 1.



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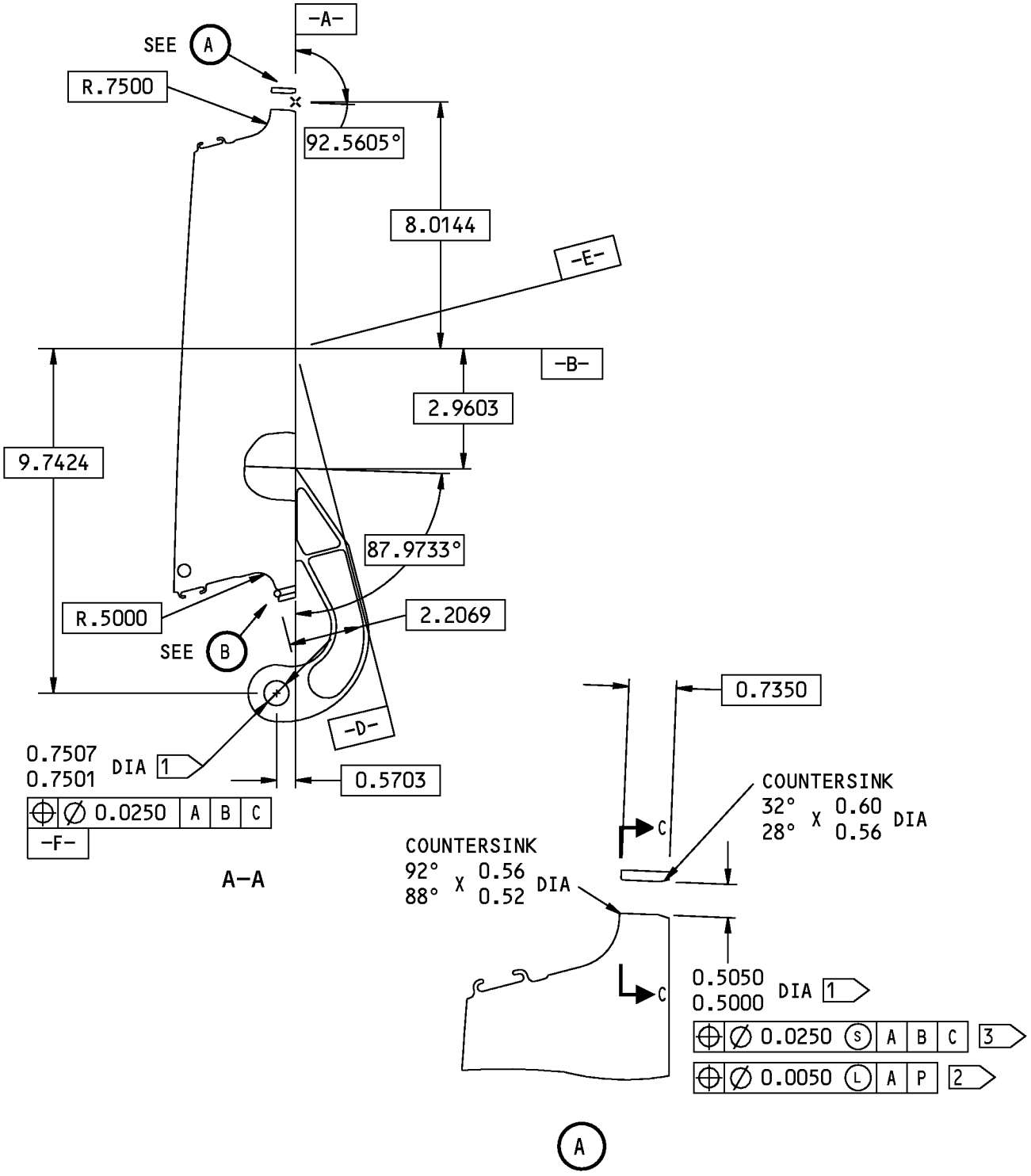
141A6804-2, 141A6816-1 Framework Repair
Figure 601 (Sheet 1 of 4)



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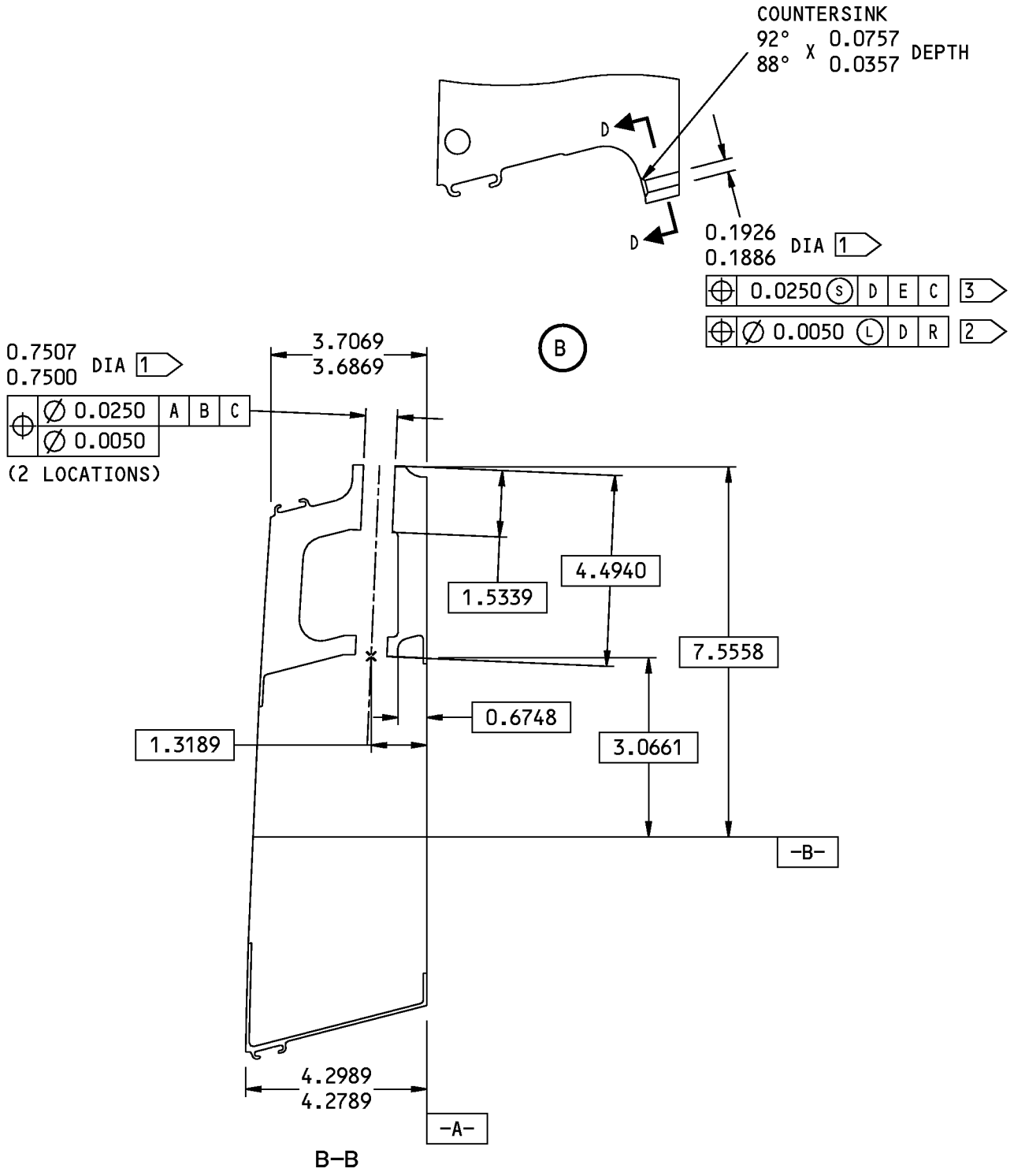
141A6804-2, 141A6816-1 Framework Repair
Figure 601 (Sheet 2 of 4)



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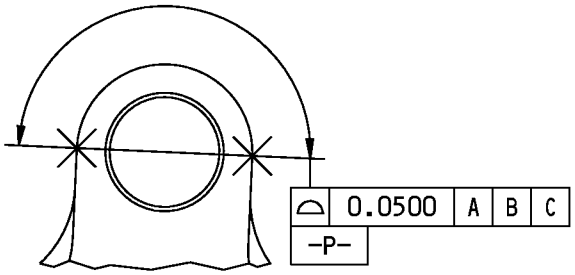
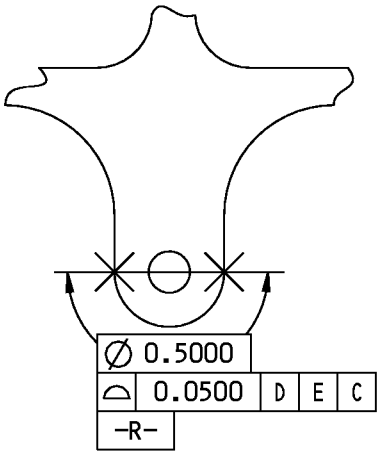
141A6804-2, 141A6816-1 Framework Repair
Figure 601 (Sheet 3 of 4)



C–C



D–D

1 DO NOT APPLY PRIMER INTO THIS HOLE

2 141A6804-2

3 141A6816-1

BREAK ALL SHARP EDGES

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

1556967 S0000287556_V1

141A6804-2, 141A6816-1 Framework Repair
Figure 601 (Sheet 4 of 4)

LATCH PIN ASSEMBLY - REPAIR 4-1

141A6810–1

1. General

A. This procedure has the data necessary to repair and refinish the latch pin assembly (120).

B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.

D. Refer to IPL Figure 1 for item numbers.

E. General repair details:

(1) Material: PH13-8MO CRES, HT TR H1000 (MIL-H-6875)

2. Bearing Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C50056	Compound - Nondrying Resin Mix Corrosion Inhibiting Material	BMS 3-27

B. References

Reference	Title
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure

NOTE: For finishing materials, refer to SOPM 20-60-02.

(1) Remove the rod end bearing (135) with the washer (130) and nut (125) from the latch pin (140).

WARNING: BMS 3-27 CORROSION INHIBITING COMPOUND CONTAINS SOLVENTS, CHROMATES, AND A SMALL AMOUNT OF BOUND ASBESTOS. CONSULT THE APPLICABLE SAFETY STANDARDS FOR APPROVED HANDLING PROCEDURES.

CAUTION: BMS 3-27 COMPOUND IS USED ONLY IN STATIC JOINTS WHERE GREASE CANNOT BE APPLIED. BMS 3-27 COMPOUND IN DYNAMIC JOINTS WILL NOT LET THEM MOVE FREELY.

(2) Apply corrosion inhibiting compound, C50056 onto the threads of the rod end bearing (135) and the nut (125).

(3) Install the new rod end bearing (135), the nut (125), and the washer (130) into the latch pin (140).

3. Pin Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
D00113	Lubricant - Liquid Dispersed Solid Film Lubricant	BMS3-8, BAC 5811, TYPE VIII

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-08	APPLICATION OF BONDED SOLID FILM LUBRICANTS
SOPM 20-60-03	LUBRICANTS

C. Procedures (REPAIR 4-1, Figure 601)

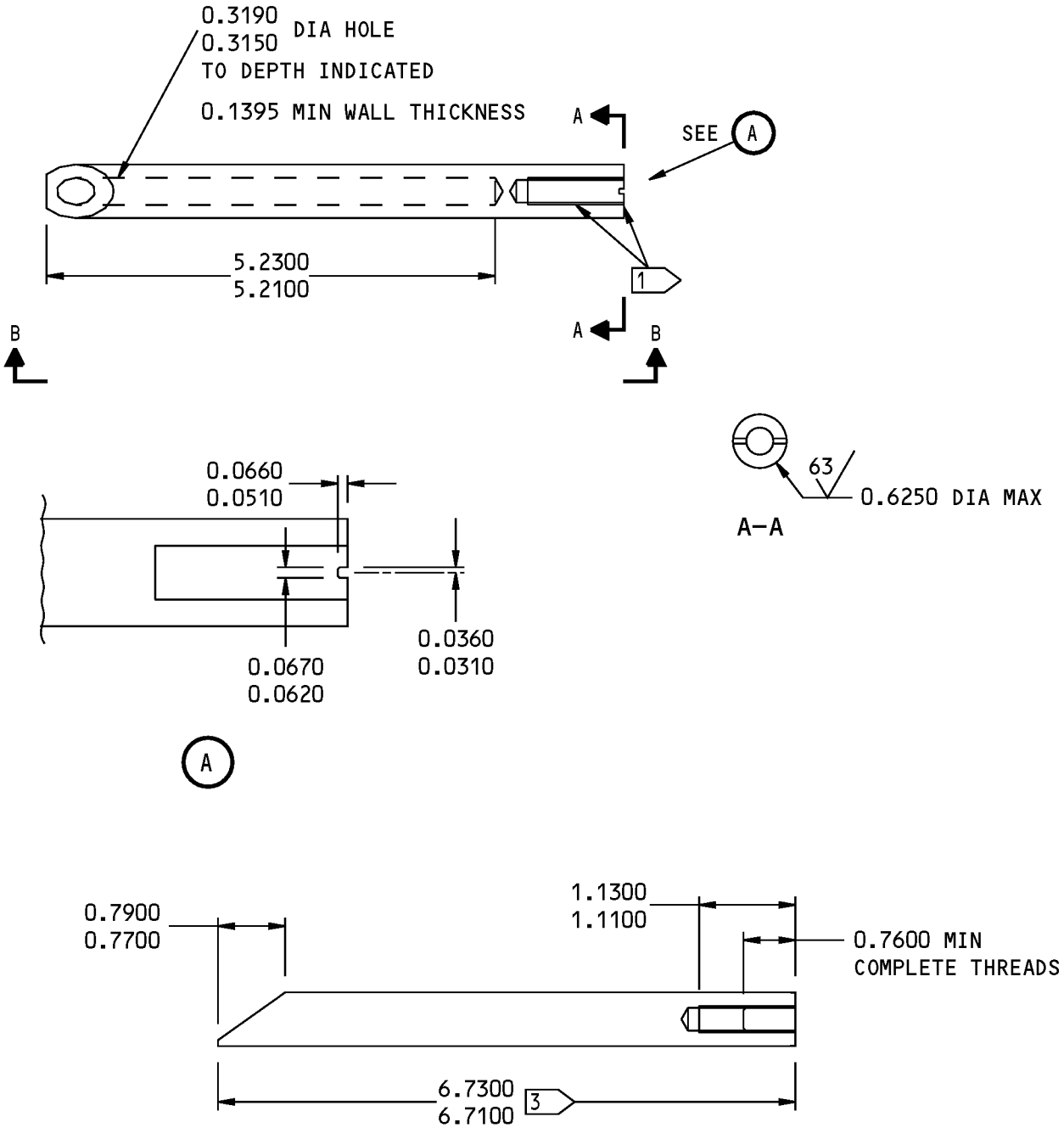
NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For the decoding table for Boeing finish codes, refer to SOPM 20-41-01. For lubricants, refer to SOPM 20-60-03.

(1) Remove protective finish from the pin (140).

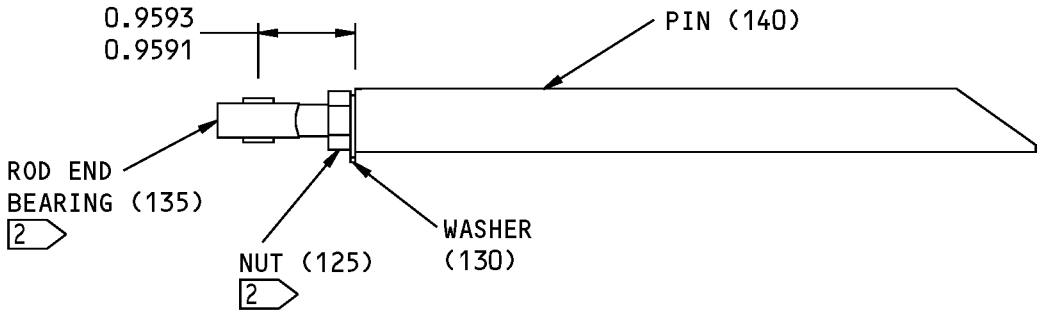
(2) Clean the pin (140).

(3) Apply cadmium plate (F-16.06) to surface identified by flagnote 1.

(4) Apply lubricant, D00113 all around the surface as shown in SOPM 20-50-08 at the location identified by flagnote 3.



141A6810-1 Latch Pin Assembly Repair
Figure 601 (Sheet 1 of 2)



141A6810-1 SHOWN

1 APPLY CADMIUM PLATE, TYPE 2, CLASS 2 (F-16.06) AS SHOWN IN SOPM 20-42-05 AT THIS LOCATION

2 APPLY BMS 3-27 CORROSION INHIBITING COMPOUND ONTO THE THREADS OF THE ROD END BEARING AND THE NUT BEFORE INSTALLATION

3 APPLY BMS 3-8 SOLID FILM LUBRICANT ALL AROUND THIS LOCATION

ITEM NUMBER REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

141A6810-1 Latch Pin Assembly Repair
Figure 601 (Sheet 2 of 2)

ASSEMBLY

1. General

A. This procedure has the data necessary to assemble the forward access door assembly.

B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

C. Refer to IPL Figure 1 for item numbers.

2. Assembly

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95
B00003	Cleaner - Emulsion Alkaline - GMC 528B	
B00052	Soap - Liquid - Turco 1526	BAC5507
B50093	Soap - Liquid - Kelite Spraywhite	BAC5507
C50056	Compound - Nondrying Resin Mix Corrosion Inhibiting Material	BMS 3-27
D00013	Grease - Aircraft And Instrument Grease	MIL-PRF-23827 (NATO G-354) (Supersedes MIL-G-23827)
G50347	Lockwire - Nickel-copper, 0.032 inch diameter	NASM20995N˜ C32

B. References

Reference	Title
SOPM 20-41-05	APPLICATION OF CORROSION INHIBITING COMPOUNDS
SOPM 20-50-01	BOLT AND NUT INSTALLATION
SOPM 20-50-02	INSTALLATION OF SAFETYING DEVICES
SOPM 20-50-05	APPLICATION OF ALUMINUM FOIL AND OTHER MARKERS
SOPM 20-60-01	CLEANING MATERIALS
SOPM 20-60-02	FINISHING MATERIALS
SOPM 20-60-03	LUBRICANTS
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. General

(1) To assemble the forward access door, it is necessary to make sure that the replaced components are installed correctly with sealant, A00247.

D. Procedure

NOTE: For bolt and nut Installation, refer to SOPM 20-50-01. For finishing materials, refer to SOPM 20-60-02. For lubricants, refer to, SOPM 20-60-03. For miscellaneous materials, refer to SOPM 20-60-04.

(1) Use standard industry procedures and the steps shown below to assemble this component.

(2) Apply fay surface seal with sealant, A00247 between the mating surface of the outer skin and the handle assembly (150) as shown in ASSEMBLY, Figure 701.

(3) Install handle assembly (150) onto the framework assembly (155) with rivets (145). Remove excess sealant, A00247 from the inside of the handle pan cutout.

(4) Assemble latch pin assembly (120) as shown in ASSEMBLY, Figure 702.

WARNING: BMS 3-27 CORROSION INHIBITING COMPOUND CONTAINS SOLVENTS, CHROMATES, AND A SMALL AMOUNT OF BOUND ASBESTOS. CONSULT THE APPLICABLE SAFETY STANDARDS FOR APPROVED HANDLING PROCEDURES.

CAUTION: BMS 3-27 COMPOUND IS USED ONLY IN STATIC JOINTS WHERE GREASE CANNOT BE APPLIED. BMS 3-27 COMPOUND IN DYNAMIC JOINTS WILL NOT LET THEM MOVE FREELY.

(a) Apply corrosion inhibiting compound, C50056 onto the threads of the rod end bearing (135) and the nut (125) (SOPM 20-41-05).

(b) Install the rod end bearing (135), the nut (125), and the washer (130) into the pin (140).

(5) Install one end of the latch linkages (120) to the handle assembly (150) with the bolt (95), the washers (100, 105), the nut (110), and cotter pin (90) as shown in ASSEMBLY, Figure 703.

(6) Lubricate the latch pin assembly (120) with grease, D00013 and install it into the framework assembly (155).

(7) Install the other end of the latch linkages (120) to the latch pin assembly (120) with the bolt (95), the washers (100, 105), the nut (110) and cotter pin (90) as shown in ASSEMBLY, Figure 703.

(8) Adjust the rod end bearing (135) to get the extended and retracted dimensions of the latch pin assembly (120) as shown in ASSEMBLY, Figure 703.

(9) Attach the nut (125) and the washer (130) with lockwire, G50347 as shown in SOPM 20-50-02.

(10) Install the drain valve retainer (60) into the framework assembly (155) with the bolt (45A), the washer (50A), and nut (55). Apply 10-15 inch-pound of torque to the nut (55) as identified by flagnote 2.

(11) Apply fay surface seal with sealant, A00247 between the mating surface of the drain valve retainer (60) and the outer skin as shown in ASSEMBLY, Figure 701. Remove excess sealant, A00247 from the drain valve cutout and drain passage.

CAUTION: DO NOT TIGHTEN THE DRAIN VALVE ASSEMBLY (65) TOO MUCH. THE HOUSING (85) FLANGE WILL CRACK OR BREAK IF TIGHTENED TOO MUCH.

(12) Install the drain valve assembly (65) from the external of the framework assembly (155) until the flange of the housing (85) is tight with the outer skin.

(13) Apply fay surface seal with sealant, A00247 between the handle support plate (30) and the handle assembly (150). Remove excess sealant, A00247 from the handle support assembly (150).

(14) Install the handle support plate (30) onto the framework assembly (155) with the washer (10) and bolt (5).

(15) Install the handle support plate (30) onto the handle assembly (150) with rivets (25).

(16) Install the seal (40B) onto the framework assembly (155).

NOTE: Use Kelite Spraywhite, B50093, Turco 1526 soap, B00052 or GMC 528B cleaner, B00003(SOPM 20-60-01) to install the seal (40) if necessary.

(a) Install the seal (40B) onto the framework assembly (155) retainer with the seal lip faced outboard and the side of the seal (40B) marked fwd-on the forward side. Begin at the framework assembly centerline and work all around.

(b) Use a circular edge to push the upper lip of the seal (40B) into the framework assembly (155) retainer.

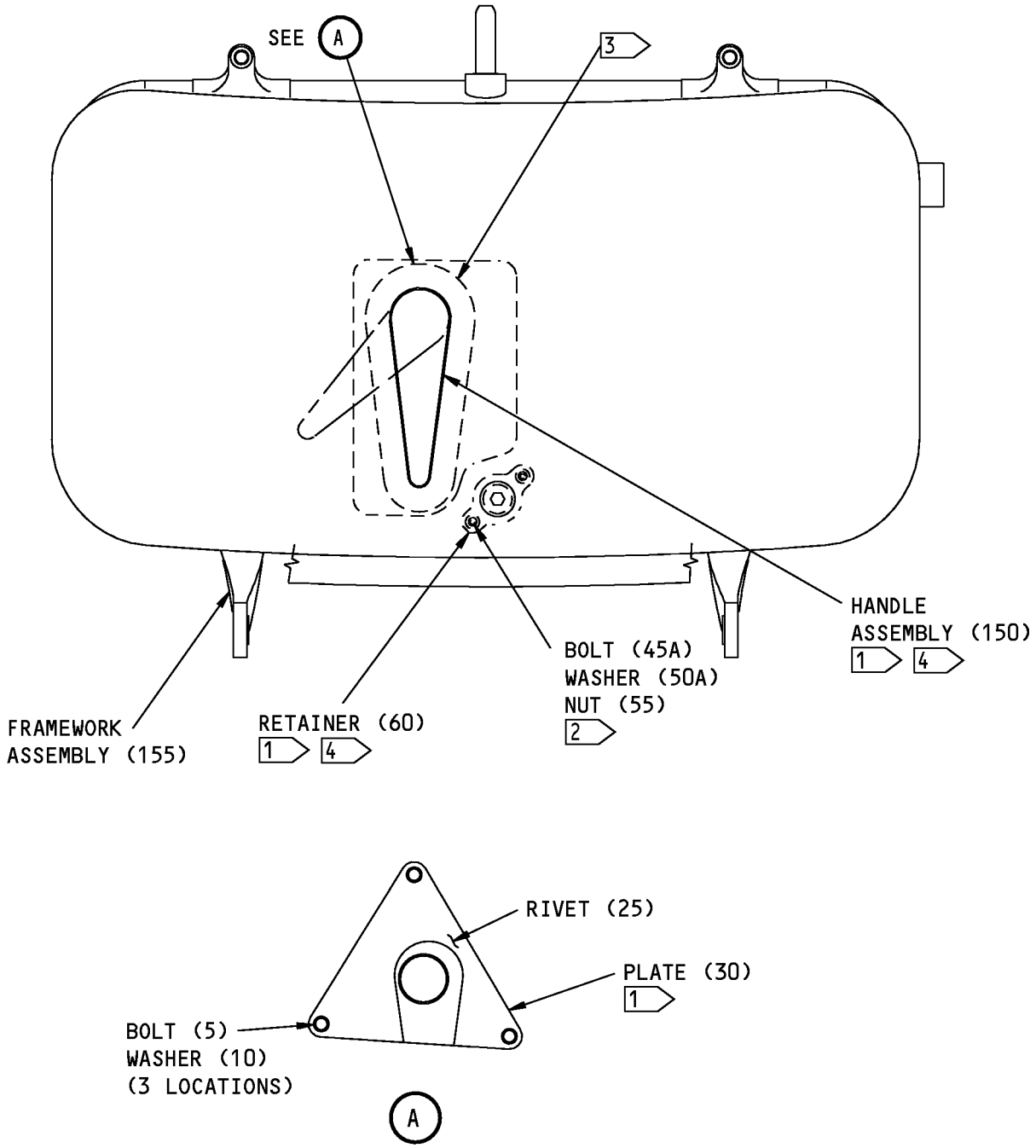
NOTE: Push perpendicular to the seal (40B) to avoid bunching or sliding.

(c) Measure the wave height from the sealing surface to the lip of the seal (40B). The maximum allowed wave height is 0.10 inches.

(d) Remove any wrinkles in the seal (40B).

NOTE: The seal (40B) is adjusted after the door is installed on the airplane.

(17) Install I.D. plate (195) and marker (200) (SOPM 20-50-05) identified by flagnotes 7 and 8 shown in ASSEMBLY, Figure 703.



G49842 S00041002522_V2

Sealant Application
Figure 701 (Sheet 1 of 2)

ITEM NUMBER REFER TO IPL FIG. 1

1 APPLY FAY SURFACE SEAL WITH BMS 5-95 SEALANT. REMOVE EXCESS SEALANT

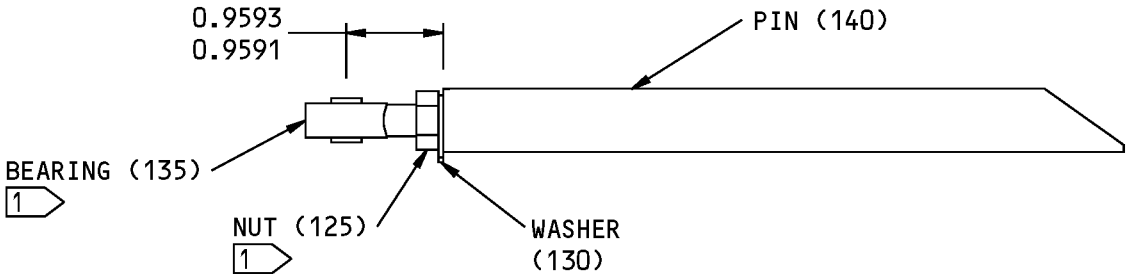
2 APPLY 10-15 INCH-POUND OF TORQUE TO NUT. NO INSPECTION REQUIRED

3 APPLY BMS 3-23 CORROSION INHIBITING COMPOUND (F-19.26) TO ALL SEAMS AND SURFACES AS SHOWN IN SOPM 20-41-05

4 DO NOT APPLY BMS 3-23 CORROSION INHIBITING COMPOUND (F-19.26) TO THIS LOCATION.

G49847 S00041002523_V2

Sealant Application
Figure 701 (Sheet 2 of 2)

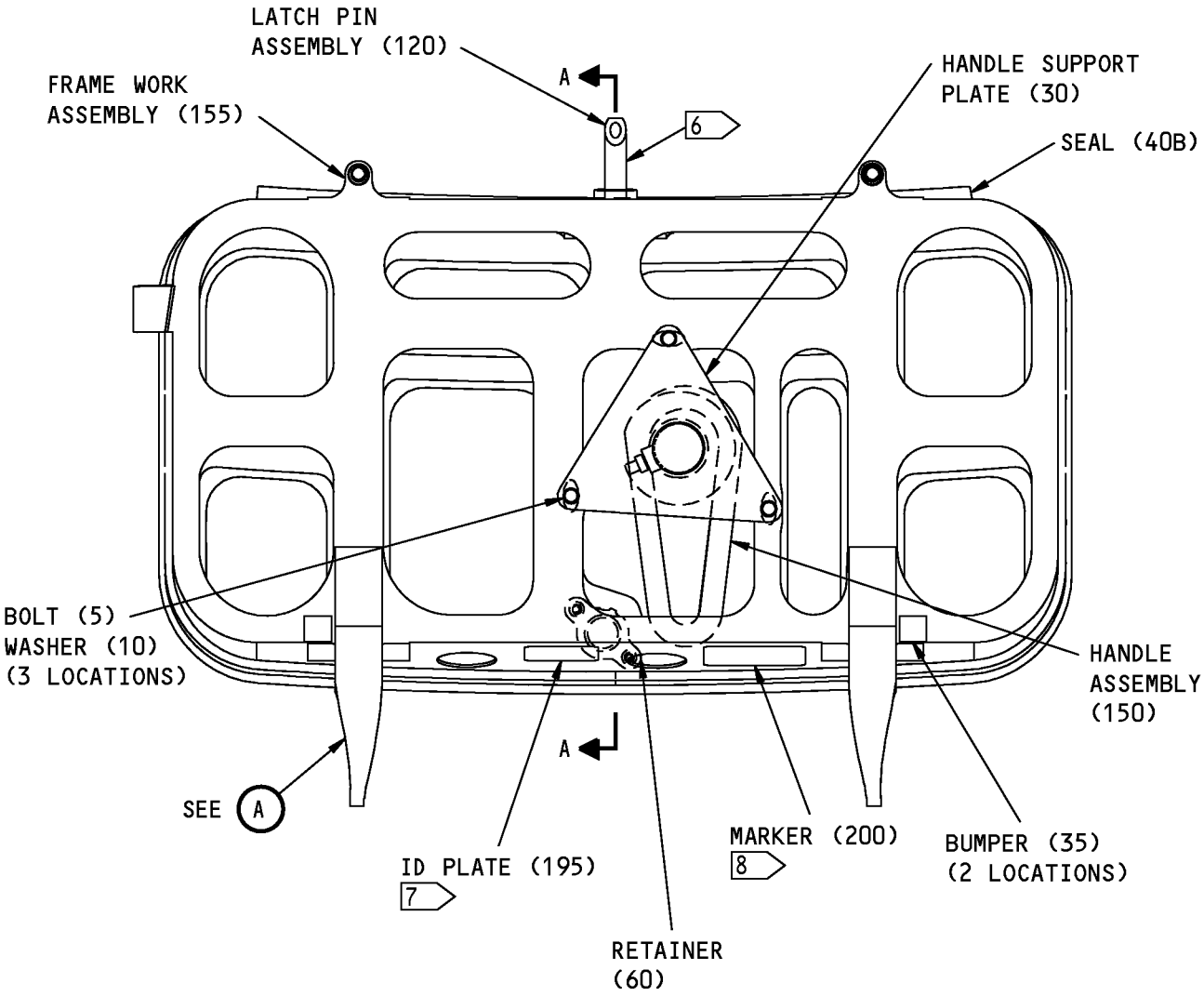


1 APPLY BMS 3-27 CORROSION INHIBITING COMPOUND ONTO THE THREADS OF THE ROD END BEARING AND NUT

ITEM NUMBER REFER TO IPL FIG. 1

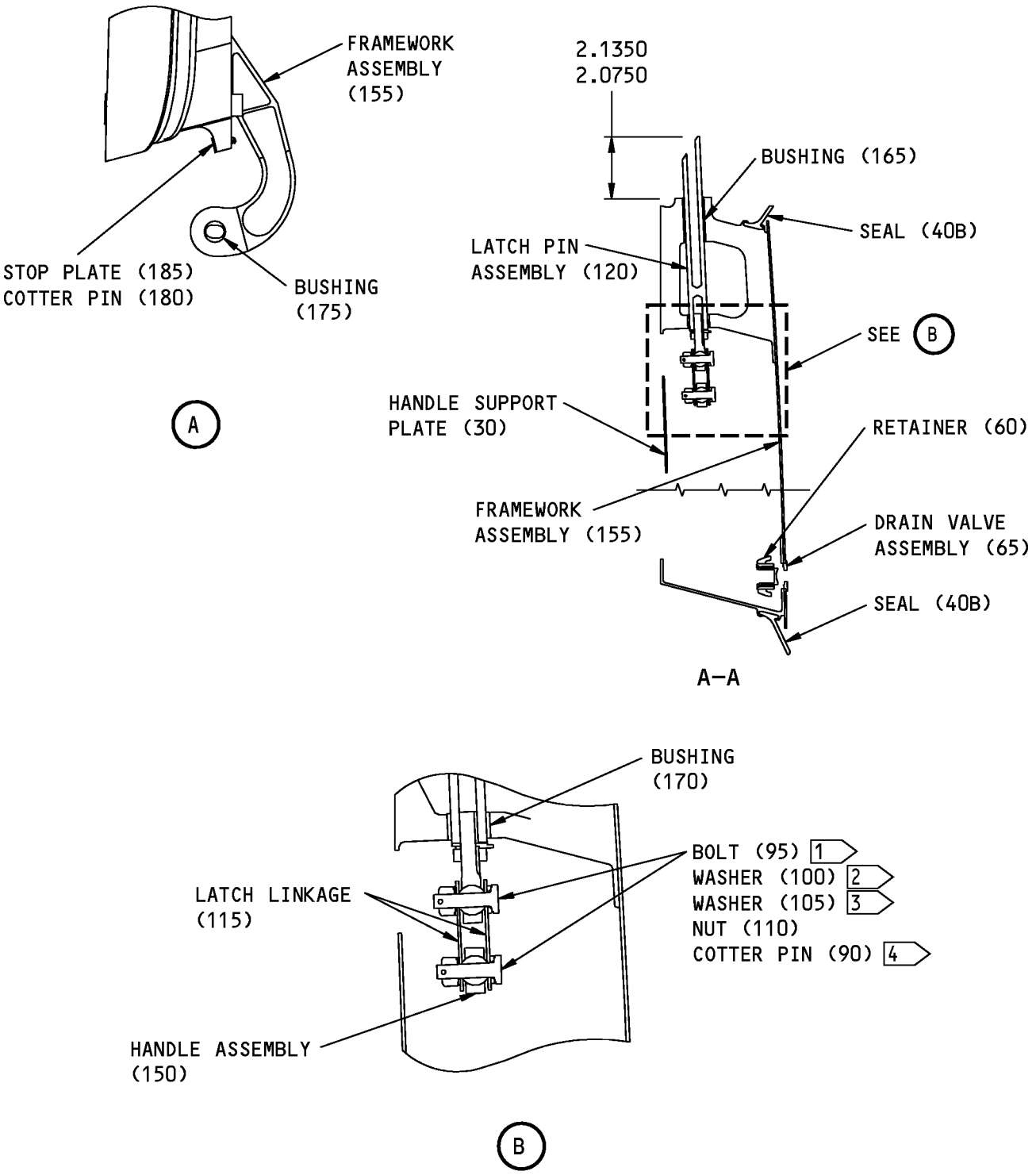
ALL DIMENSIONS ARE IN INCHES

Latch Pin Assembly
Figure 702



G49862 S00041002525_V2

Forward Access Door Assembly
Figure 703 (Sheet 1 of 3)



G49870 S00041002526_V2

Forward Access Door Assembly
Figure 703 (Sheet 2 of 3)

1 > DIRECTION OF BOLT IS OPTIONAL

2 > INSTALL UNDER BOLT HEAD

3 > INSTALL UNDER NUT

4 > INSTALL AS SHOWN IN SOPM 20-50-02

5 > INSTALL MS20995NC32 LOCKWIRE AT THIS LOCATION

6 > APPLY MIL-PRF-23827 GREASE AT THIS LOCATION

7 > INSTALL ID PLATE AS SHOWN IN SOPM 20-50-05 AT THIS LOCATION

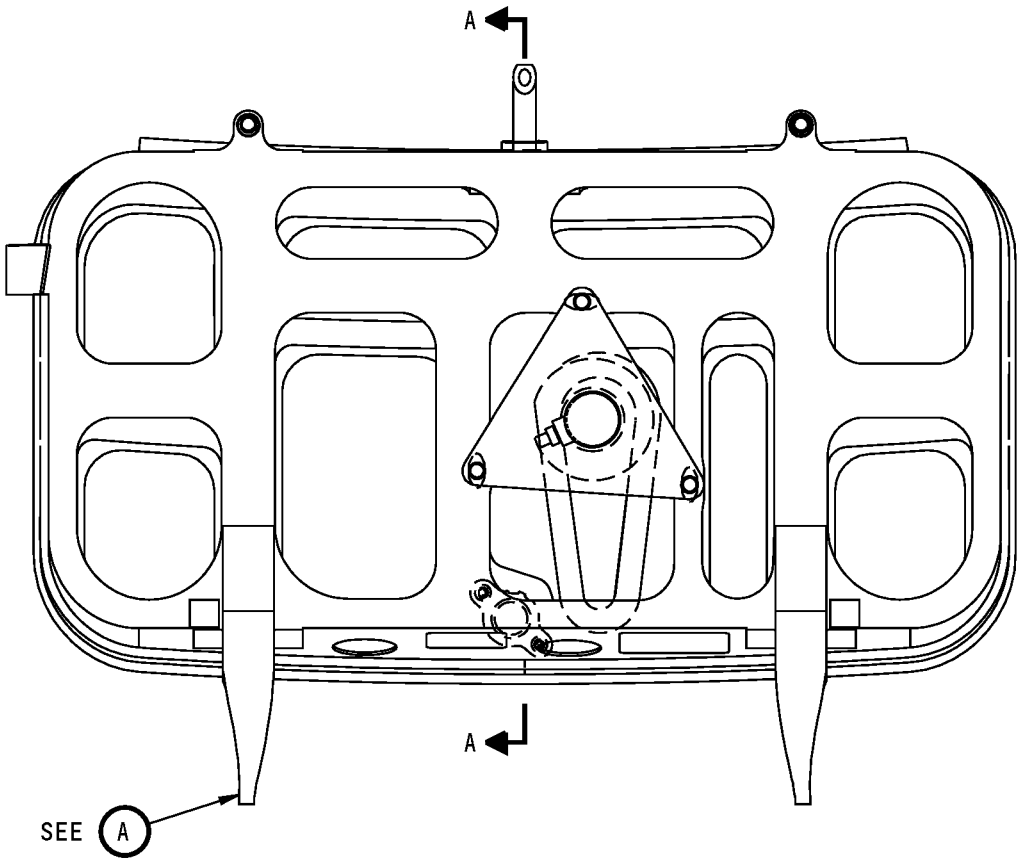
8 > INSTALL MARKER AS SHOWN IN SOPM 20-50-05 AT THIS LOCATION

ITEM NUMBER REFER TO IPL FIG. 1

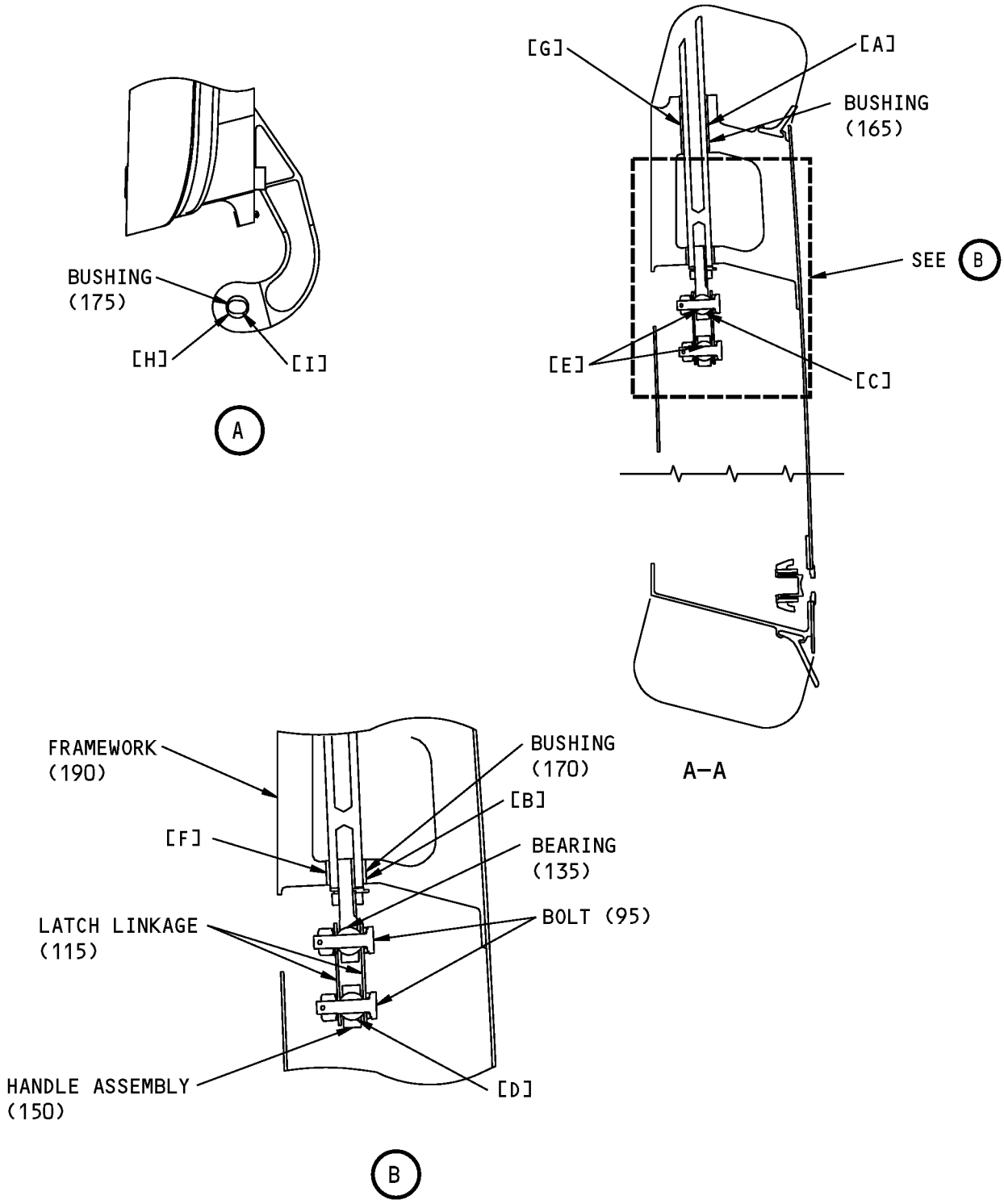
ALL DIMENSIONS ARE IN INCHES

G55313 S00041002527_V2

Forward Access Door Assembly
Figure 703 (Sheet 3 of 3)



Fits and Clearances
Figure 801 (Sheet 1 of 3)



Fits and Clearances
Figure 801 (Sheet 2 of 3)

REF LETTER	REF IPL FIG. 1, MATING ITEM NO.	DESIGN DIMENSION*				SERVICE WEAR LIMIT*		
		DIMENSION		ASSEMBLY CLEARANCE 1		DIMENSION		MAXIMUM CLEARANCE
		MIN	MAX	MIN	MAX	MIN	MAX	
[A]	ID 165	0.6250	0.6265	0.0000	0.0016		0.6281	0.0032
	OD 140	0.6249	0.6250			0.6249		
[B]	ID 170	0.6250	0.6265	0.0000	0.0016		0.6281	0.0032
	OD 140	0.6249	0.6250			0.6249		
[C]	ID 135	0.2495	0.2500	0.0000	0.0019		0.2519	0.0043
	OD 95	0.2481	0.2495			0.2476		
[D]	ID 150	0.2495	0.2500	0.0000	0.0019		0.2519	0.0043
	OD 95	0.2481	0.2495			0.2476		
[E]	ID 115	0.2500	0.2540	0.0005	0.0059		0.2599	0.0177
	OD 95	0.2481	0.2495			0.2422		
[F]	ID 190	0.7500	0.7507	−0.0019	−0.0007		0.7500	0.0000
	OD 170	0.7514	0.7519			0.7500		
[G]	ID 190	0.7500	0.7507	−0.0019	−0.0007		0.7500	0.0000
	OD 165	0.7514	0.7519			0.7500		
[H]	ID 190	0.7501	0.7506	−0.0017	−0.0006		0.7501	0.0005
	OD 175	0.7512	0.7518			0.7506		
[I]	ID 175	0.4375	0.4380	0.0005	0.0015		0.4395	0.0035
	OD 2	0.4365	0.4370			0.4360		

* ALL DIMENSIONS ARE IN INCHES

1 NEGATIVE VALUES INDICATE INTERFERENCE FIT

2 INSTALLATION BUSHING BACB28AK05-068

Fits and Clearances
Figure 801 (Sheet 3 of 3)

(NOT APPLICABLE)

ILLUSTRATED PARTS LIST

1. Introduction

A. The Illustrated Parts List (IPL) contains an illustration and a list of component parts you can repair or replace. The Illustrated Parts Catalog (IPC) shows how to use the Boeing part number system.

B. This shows how parts are related: The relation of each item to its next higher assembly (NHA) is shown in the NOMENCLATURE column. Use the indenture system that follows:

1 2 3 4 5 6 7

. Assembly

. Attaching parts for assembly

. . Detail parts for assembly

. . Subassembly

. . Attaching parts for subassembly

. . . Detail parts for subassembly

. . . Sub-subassembly

. . . Attaching parts for subassembly

. . . . . Details parts for sub-subassembly

Detail Installation Parts (Included only if installation parts may be sent to the shop as part of assembly)

C. Each top assembly is given one use code letter (A, B, C, etc.) in the USAGE CODE column. All subsequent component parts in the list can have one or more of the use code letters to show effectivity to top assemblies. A component part without a use code applies to all top assemblies.

D. An alphabetical letter is added after the item number for optional parts, parts changed by a Service Bulletin, configuration differences (except left-handed and right-handed parts), last engineering releases, and parts added between item numbers in a sequence. The alphabetical letter will not be shown on the illustration for equivalent parts of the same part number.

E. Color-coded parts are identified with a single digit alpha following the dash number or with ''SP'' suffix. If the ''SP'' suffix is used, it represents consolidation of all color codes applicable for a given usage which are not separately listed. Orders for color-coded parts should include the registry number of the airplane for which the parts are ordered.

F. If a part number is 15 characters long but will not fit in the part number column, the part number will be displayed with a ''~'' at the end of the line and will be continued on the next line. The ''~'' denotes that the part number continues on the next line.

G. Parts changed by a Service Bulletin are shown by PRE SB XXXX and POST SB XXXX added to the NOMENCLATURE column.

(1) When a new top assembly is added by a Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the top assembly level only. The configuration differences at the detail part level are shown by use code letters.

(2) When the top assembly part number is not changed by the Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the detail level.

H. Interchangeable Parts

Optional (OPT)	The part is optional to and interchangeable with other parts that have the same item number.
Replaces, Replaced by and not interchangeable with (REPLACES, REPLACED BY AND NOT INTCHG/W)	The part replaces and is not interchangeable with the initial part.
Replaces, Replaced by (REPLACES, REPLACED BY)	The part replaces and is interchangeable with, or is an alternative to, the initial part.

VENDOR CODES

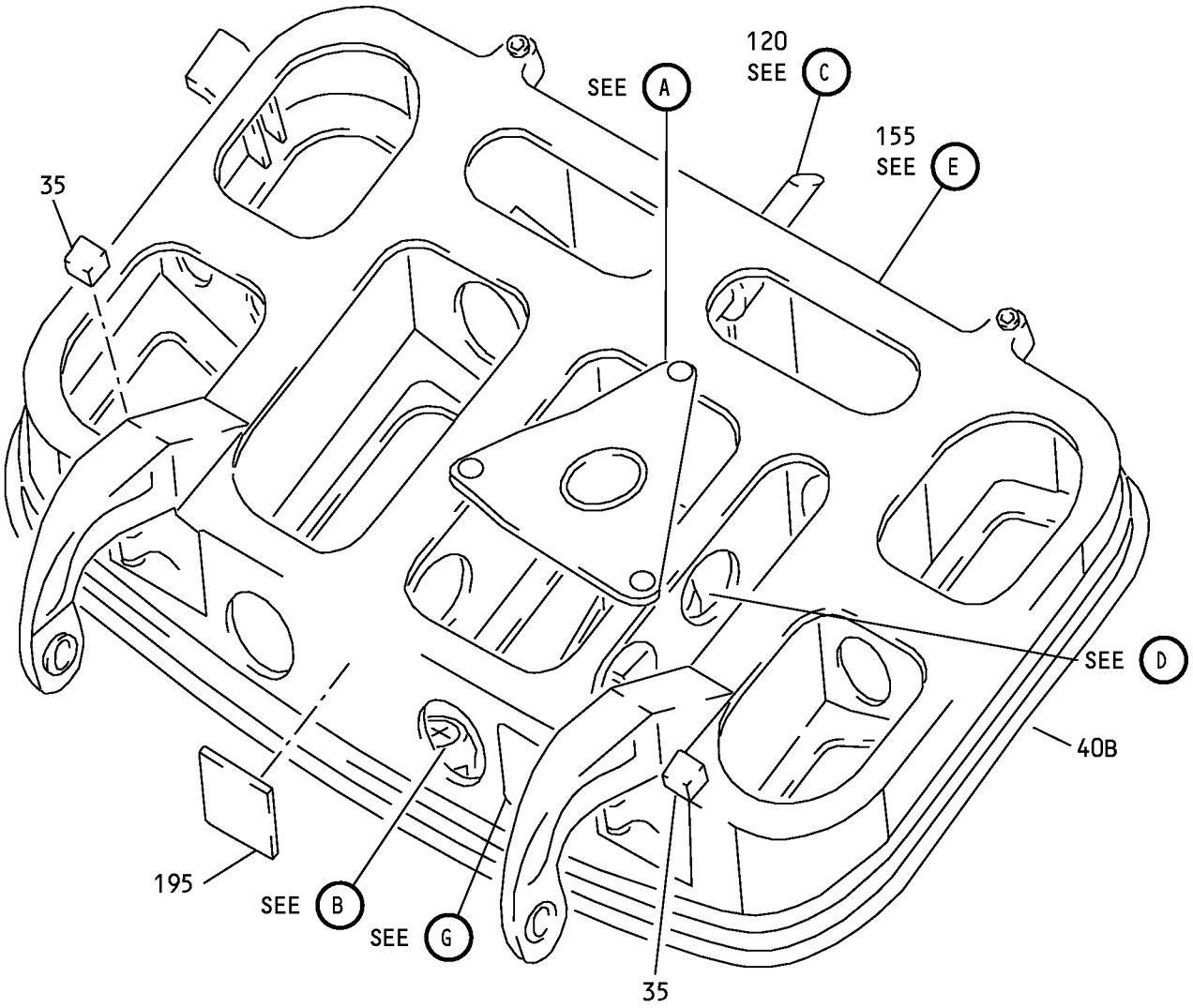
Code	Name
11815	CHERRY AEROSPACE FASTENERS DIV OF TEXTRON 1224 EAST WARNER AVENUE PO BOX 2157 SANTA ANA, CALIFORNIA 92707-0157 FORMERLY IN LOS ANGELES, CALIF , FORMERLY CHERRY FASTENERS TOWNSEND DIV OF TEXTRON INC V71087
15653	ALCOA GLOBAL FASTENERS INC DIV KAYNAR PRODUCTS 800 S STATE COLLEGE BLVD FULLERTON, CALIFORNIA 92831-3001 FORMERLY VK6405 MICRODOT AEROSP LTD; FORMERLY KAYNAR TECH FORMERLY FAIRCHILD FASTENERS KAYNAR DIV
52828	REPUBLIC FASTENER MFG CORP 1300 RANCHO CONEJO BLVD NEWBURY PARK, CALIFORNIA 91320-1405 FORMERLY IN SYLMAR, CALIFORNIA
53551	ALLFAST FASTENING SYSTEMS INC 15200 EAST DON JULIAN ROAD PO BOX 3166 CITY OF INDUSTRY, CALIFORNIA 91745-1001 FORMERLY V0736B FORMERLY ALLFAST INC V5K545
62554	SIMMONDS MECAERO FASTENERS INC 1734 SEQUOIA AVENUE ORANGE, CALIFORNIA 92668

Code	Name
72962	HARVARD INDUSTRIES INC 3 WERNER WAY SUITE 210 LEBANON, NEW JERSEY 08833 FORMERLY ESNA V7A079 FORMERLY ELASTIC STOP NUT IN UNION, NJ
80539	SPS TECHNOLOGIES INC DIV AERPSOACE - SANTA ANA 2701 SOUTH HARBOR BOULEVARD SANTA ANA, CALIFORNIA 92704-5803 FORMERLY NUTT-SHEL DIV OF SPC WESTERN CO V80539 AND STANDARD PRESSED STEEL WESTERN DIV V17279
83014	HARTWELL CORPORATION 900 SOUTH RICHFIELD ROAD PLACENTIA, CALIFORNIA 92670-6732 FORMERLY V0532B IN LOS ANGELES, CALIFORNIA
98996	OLYMPIC FASTENING SYSTEMS INC DOWNEY, CALIFORNIA 90241-4986 OBSOLETE RECORD
U1901	BESTOBEL AVIATION PART MEGGITT COMPOSITES 127/135 FARNHAM ROAD SLOUGH BERKS SLI 4UY UNITED KINGDOM FORMERLY MEGGITT AEROSP COMPTNS BESTOBELL PROD FORMERLY VK8319

NUMERICAL INDEX

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
10-60476-14		1	40B	1
102F9201-3		1	20	3
140N2020-1		1	80	1
140N2021-1		1	60	1
140N2022-1		1	65	1
140N2022-2		1	85	1
140N2022-3		1	75	1
140N2022-4		1	70	1
141A6802-3		1	1A	RF
141A6802-4		1	1B	RF
141A6802-5		1	1C	RF
141A6802-6		1	1D	RF
141A6803-2		1	155	1
141A6803-3		1	155A	1
141A6804-2		1	190	1
141A6805-1		1	30	1
141A6806-1		1	35	2
141A6807-1		1	115	2
141A6808-2		1	185	2
141A6809-1		1	175	2
141A6809-2		1	175A	2
141A6810-1		1	120	1
141A6810-2		1	140	1
141A6815-1		1	155B	1
141A6816-1		1	190A	1
66-12688-13		1	160	2
AF5141-3C		1	15	6
BA205195		1	40B	1
BAC27DBY191		1	200	1
BACB28AW10B046C		1	170	1
BACB28AW10B155C		1	165	1
BACB30NR4DK10		1	95	2
BACB30NT3K4		1	5	3
BACB30VF08K3		1	45A	2

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
BACN10JD4CD		1	110	2
BACN10JR3CFD		1	20	3
BACN10YR08CD		1	55	2
BACP18BC02A06P		1	90	2
BACR15DR3AC		1	15	6
BACR15FT5D		1	25	8
BACR15GF5D		1	145	24
BACW10BN3AC		1	10	3
BACW10BN4AC		1	100	2
BRF200C3D		1	20	3
CCR264CS3C		1	15	6
H52732-08CD		1	55	2
HA252-1		1	150	1
K51602-3BAC		1	20	3
M81935-1-4K		1	135	1
MS24665-82		1	180	2
MS27253F3		1	195	1
NAS1149D0432J		1	105	2
NAS1149DN832J		1	50A	2
NAS509-5		1	125	1
NAS513-5		1	130	1
NS202476-02		1	20	3
PLH508CD		1	55	2
RV541A3C		1	15	6
S141A681-1		1	150	1
T8092C1032CD		1	20	3



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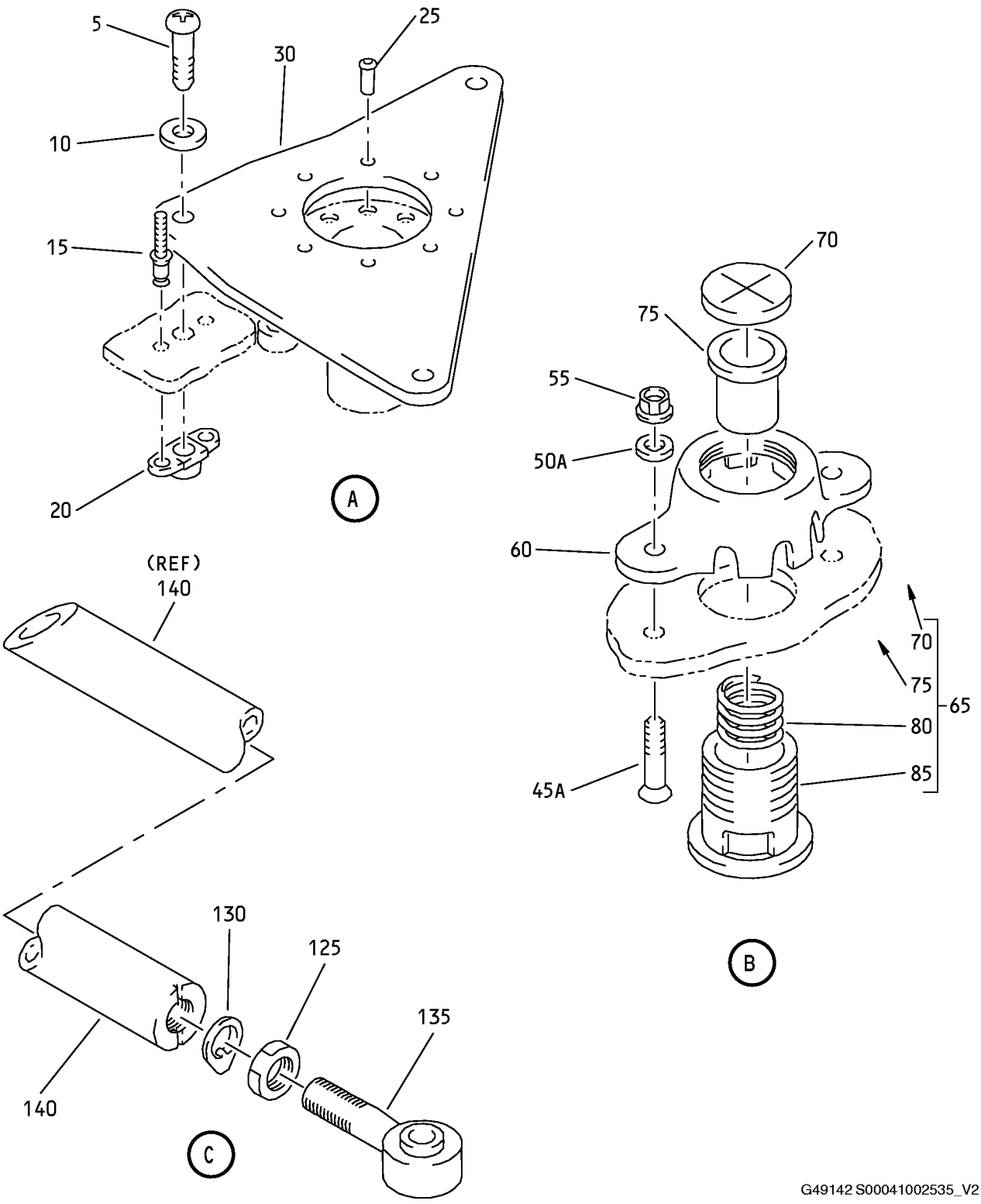
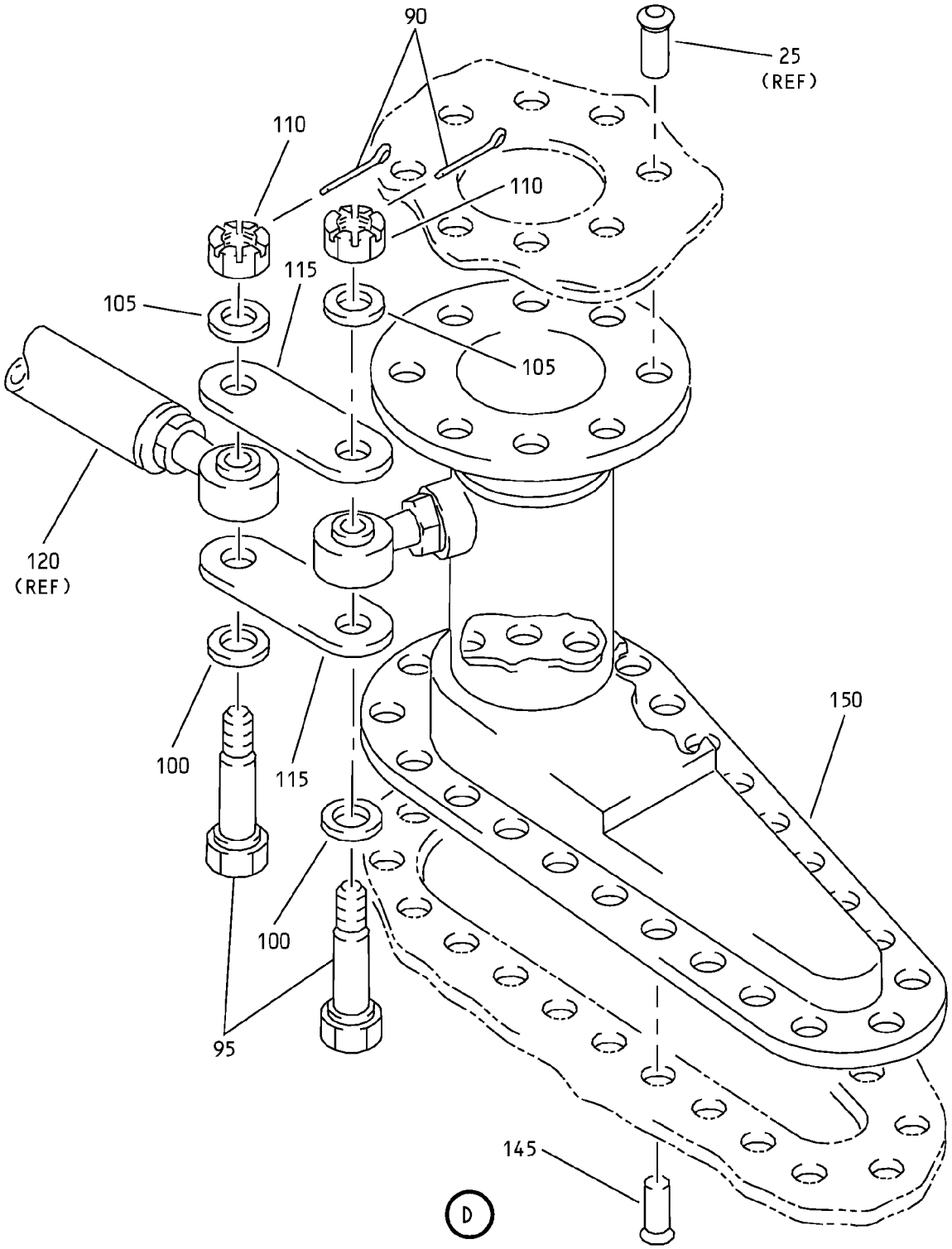
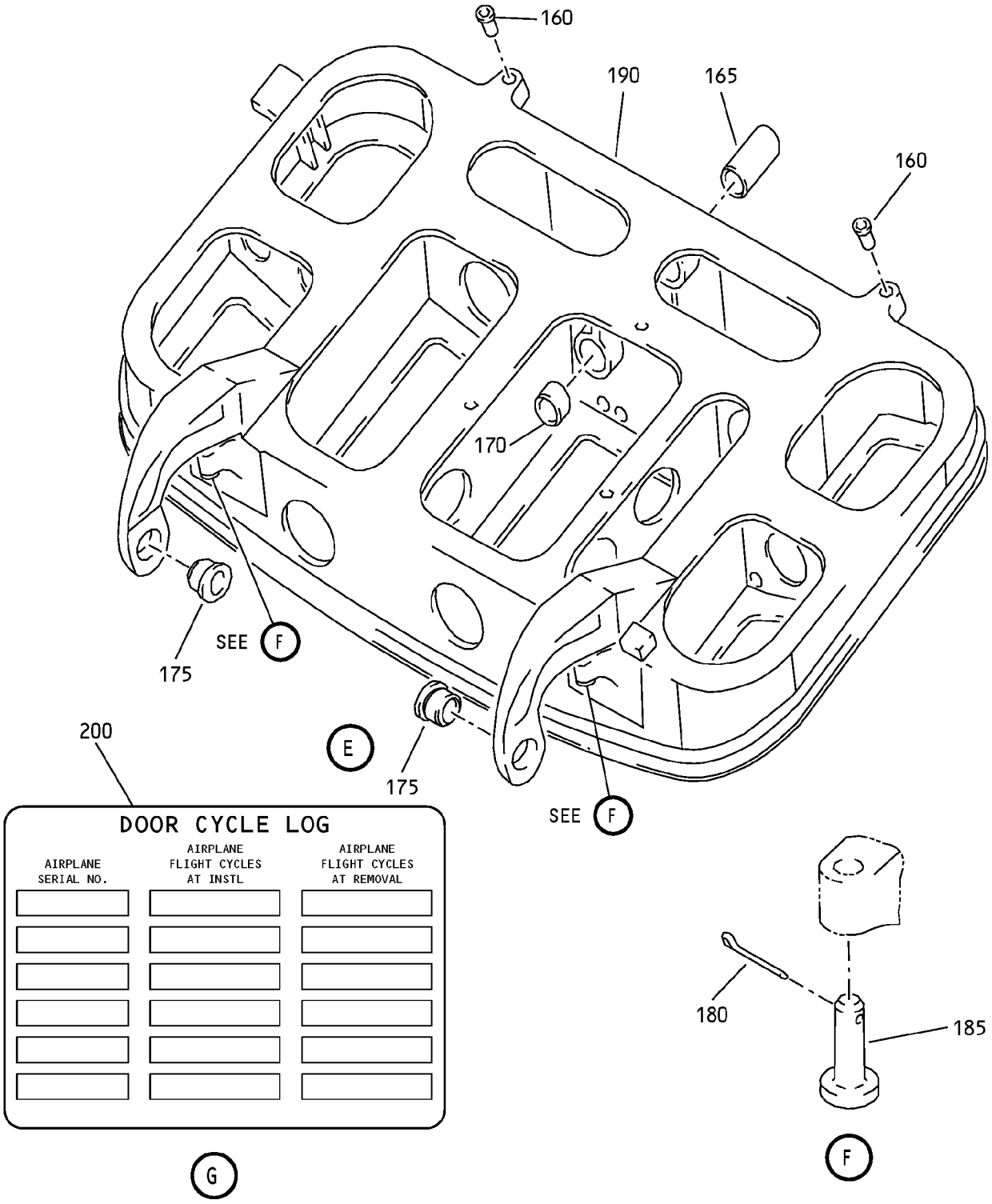
Forward Access Door Assembly
IPL Figure 1 (Sheet 1 of 4)



Forward Access Door Assembly
IPL Figure 1 (Sheet 2 of 4)



Forward Access Door Assembly
IPL Figure 1 (Sheet 3 of 4)



Forward Access Door Assembly
IPL Figure 1 (Sheet 4 of 4)

FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USAGE CODE	UNITS PER ASSY
1–					
–1A	141A6802-3		DOOR ASSY-FWD ACCESS. PNL NO. 112A	A	RF
–1B	141A6802-4		DOOR ASSY-FWD ACCESS. PNL NO. 112A	B	RF
–1C	141A6802-5		DOOR ASSY-FWD ACCESS. PNL NO. 112A	C	RF
–1D	141A6802-6		DOOR ASSY-FWD ACCESS. PNL NO. 112A	D	RF
5	BACB30NT3K4		. BOLT		3
10	BACW10BN3AC		. WASHER		3
15	AF5141-3C		. RIVET (V53551) (SPEC BACR15DR3AC) (OPT CCR264CS3C (V11815)) (OPT RV541A3C (V98996)) (SIZE DETERMINED ON INST)		6
20	BRF200C3D		. NUTPLATE (V52828) (SPEC BACN10JR3CFD) (OPT K51602-3BAC (V15653)) (OPT NS202476-02 (V80539)) (OPT 102F9201-3 (V72962)) (OPT T8092C1032CD (V11815))		3
25	BACR15FT5D		. RIVET (SIZE DETERMINED ON INST)		8
30	141A6805-1		. PLATE-HANDLE SPRT		1
35	141A6806-1		. BUMPER		2
40	BA201666		DELETED		
40A	10-60476-14		DELETED		
40B	BA205195		. SEAL (VU1901) (SPEC 10-60476-14)		1
–40C	9577-14		DELETED		
45	BACB30VU5HK4		DELETED		
45A	BACB30VF08K3		. BOLT		2
50	NAS1149D0332J		DELETED		
50A	NAS1149DN832J		. WASHER		2

–Item not Illustrated

FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USAGE CODE	UNITS PER ASSY
1–					
55	H52732-08CD		. NUT (V15653) (SPEC BACN10YR08CD) (OPT PLH508CD (V62554))		2
60	140N2021-1		. RETAINER-DRAIN VALVE		1
65	140N2022-1		. VALVE ASSY-DRAIN		1
70	140N2022-4		. . CAP		1
75	140N2022-3		. . PLUNGER		1
80	140N2020-1		. . SPRING		1
85	140N2022-2		. . HOUSING		1
90	BACP18BC02A06P		. PIN-COTTER		2
95	BACB30NR4DK10		. BOLT		2
100	BACW10BN4AC		. WASHER		2
105	NAS1149D0432J		. WASHER		2
110	BACN10JD4CD		. NUT		2
115	141A6807-1		. LINKAGE-LATCH		2
120	141A6810-1		. PIN ASSY-LATCH		1
125	NAS509-5		. . NUT		1
130	NAS513-5		. . WASHER		1
135	M81935-1-4K		. . BEARING		1
140	141A6810-2		. . PIN		1
145	BACR15GF5D		. RIVET (SIZE DETERMINED ON INST)		24
150	HA252-1		. HANDLE ASSY (V83014) (SPEC S141A681-1)		1
155	141A6803-2		. FRAMEWORK ASSY	A, B	1
–155A	141A6803-3		. FRAMEWORK ASSY (OPT ITEM 155B)	C, D	1
–155B	141A6815-1		. FRAMEWORK ASSY (OPT ITEM 155A)	C, D	1
160	66-12688-13		. . BUSHING-STOP		2
165	BAC˜ B28AW10B155C		. . BUSHING		1

–Item not Illustrated

FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE 1 2 3 4 5 6 7	USAGE CODE	UNITS PER ASSY
1–					
170	BAC˜ B28AW10B046C		. . BUSHING		1
175	141A6809-1		. . BUSHING-HINGE	A, B	2
–175A	141A6809-2		. . BUSHING-HINGE	C, D	2
180	MS24665-82		. . PIN-COTTER		2
185	141A6808-2		. . PLATE-STOP		2
190	141A6804-2		. . FRAMEWORK (USED ON ITEMS 155, 155A)		1
–190A	141A6816-1		. . FRAMEWORK (USED ON ITEM 155B)		1
195	MS27253F3		. PLATE		1
200	BAC27DBY191		. MARKER-ALUMINUM FOIL		1

–Item not Illustrated