



COMPONENT MAINTENANCE MANUAL WITH ILLUSTRATED PARTS LIST

WING LEADING EDGE NO. 1 AND 8 SLAT ASSEMBLY

PART NUMBER

114A5010-1, -10, -11, -12, -2, -201, -202, -3, -4, -9

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COMPONENT MAINTENANCE MANUAL

Revision No. 13
Jul 01/2009

To: All holders of WING LEADING EDGE NO. 1 AND 8 SLAT ASSEMBLY 57-43-01.

Attached is the current revision to this COMPONENT MAINTENANCE MANUAL

The COMPONENT MAINTENANCE MANUAL is furnished either as a printed manual, on microfilm, or digital products, or any combination of the three. This revision replaces all previous microfilm cartridges or digital products. All microfilm and digital products are reissued with all obsolete data deleted and all updated pages added.

For printed manuals, changes are indicated on the List of Effective Pages (LEP). The pages which are revised will be identified on the LEP by an R (Revised), A (Added), O (Overflow, i.e. changes to the document structure and/or page layout), or D (Deleted). Each page in the LEP is identified by Chapter-Section-Subject number, page number and page date.

Pages replaced or made obsolete by this revision should be removed and destroyed.

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114A5010



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Location of Change

Description of Change

NO HIGHLIGHTS

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HIGHLIGHTS

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A = Added, R = Revised, D = Deleted, O = Overflow

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TEMPORARY REVISION AND SERVICE BULLETIN RECORD

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVE	DATE OF INCORPORATION INTO MANUAL
		PRR38040-19	SEP 01/97
		PRR 38157	MAR 01/03

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TR AND SB RECORD

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All revisions to this manual will be accompanied by transmittal sheet bearing the revision number. Enter the revision number in numerical order, together with the revision date, the date filed and the initials of the person filing.

Revision		Filed	
Number	Date	Date	Initials

Revision		Filed	
Number	Date	Date	Initials



COMPONENT MAINTENANCE MANUAL

All temporary revisions to this manual will be accompanied by a cover sheet bearing the temporary revision number. Enter the temporary revision number in numerical order, together with the temporary revision date, the date the temporary revision is inserted and the initials of the person filing. When the temporary revision is incorporated or cancelled, and the pages are removed, enter the date the pages are removed and the initials of the person who removed the temporary revision.

Temporary Revision		Inserted		Removed		Temporary Revision		Inserted		Removed	
Number	Date	Date	Initials	Date	Initials	Date	Initials	Number	Date	Date	Initials

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COMPONENT MAINTENANCE MANUAL

INTRODUCTION

1. General

- A. The instructions in this manual supply the data necessary to do the maintenance functions together with the test, fault isolation, repair, and replacement of the defective parts.
- B. This manual is divided into different parts:
 - (1) Title Page
 - (2) Transmittal Letter
 - (3) Highlights
 - (4) List of Effective Pages
 - (5) Table of Contents
 - (6) Temporary Revision & Service Bulletin Record
 - (7) Record of Revisions
 - (8) Record of Temporary Revisions
 - (9) Introduction
 - (10) Procedures & IPL Sections
- C. Components that can be repaired have a different repair number for each specified repair. To find the repair number location of a component, look in the Repair-General procedure at the beginning of the REPAIR section. The Repair-General procedure also has an explanation of the True Position Dimension symbols used.
- D. All dimensions, measures, quantities and weights included are in English units. When metric equivalents are given they will be in the parentheses that follow the English units.
- E. The introduction to the Illustrated Parts List (IPL) shows how the IPL data is used.
- F. Design changes, optional parts, configuration differences and Service Bulletin modifications may cause different part numbers. These part numbers are identified in the IPL with an alphabetical letter which is added to the end of the basic item number. This new item number is referred to as an alpha-variant. Throughout the manual, IPL basic item number references also apply to alpha-variants unless shown differently.
- G. The tool reference numbers found in the individual procedures and in the Special Tools, Fixtures, and Equipment section are used to identify if a tool is a standard tool (STD-XXXX), a commercial tool (COM-XXXX), or a Special Tool (SPL-XXXX). This reference number is also used to distinguish between tools with similar names in the same procedure. These reference numbers are for use in the documentation only. They are not to be used for ordering tools.

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INTRODUCTION

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WING LEADING EDGE NO. 1 AND 8 SLAT ASSEMBLIES - DESCRIPTION AND OPERATION

1. Description

A. There are four leading edge slat assemblies installed on each wing outboard of the engine. The slat assemblies are numbered 1 through 8 from the left wing to the right wing. The slat assembly consists of aluminum alloy ribs, inner and outer clad aluminum skins, and a honeycomb trailing edge wedge. Each slat is supported by two main tracks and two auxiliary arm rib assemblies which ride on rollers in the wing leading edge. The main tracks are made of steel and the auxiliary arm rib assemblies are made of aluminum alloy. The auxiliary arm rib assemblies are part of the slat rib. The space between the inner and outer skins provides a path for the thermal anti-icing (TAI) ducts. The TAI ducts installed in the slat leading edge connect with hot air supply lines through a telescoping tube.

2. Operation

A. The lift capability of the wings is supplemented by the leading edge slat assemblies. The leading edge slat assemblies in coordination with the trailing edge flap assemblies improve the operation of the airplane at low speeds and at minimum distance takeoffs.

3. Leading Particulars (Approximate)

- A. Length – 134 inches
- B. Width – 17 inches (main track assemblies not installed) 30 inches (with the main track assemblies installed)
- C. Height – 7 inches
- D. Weight – 78 pounds

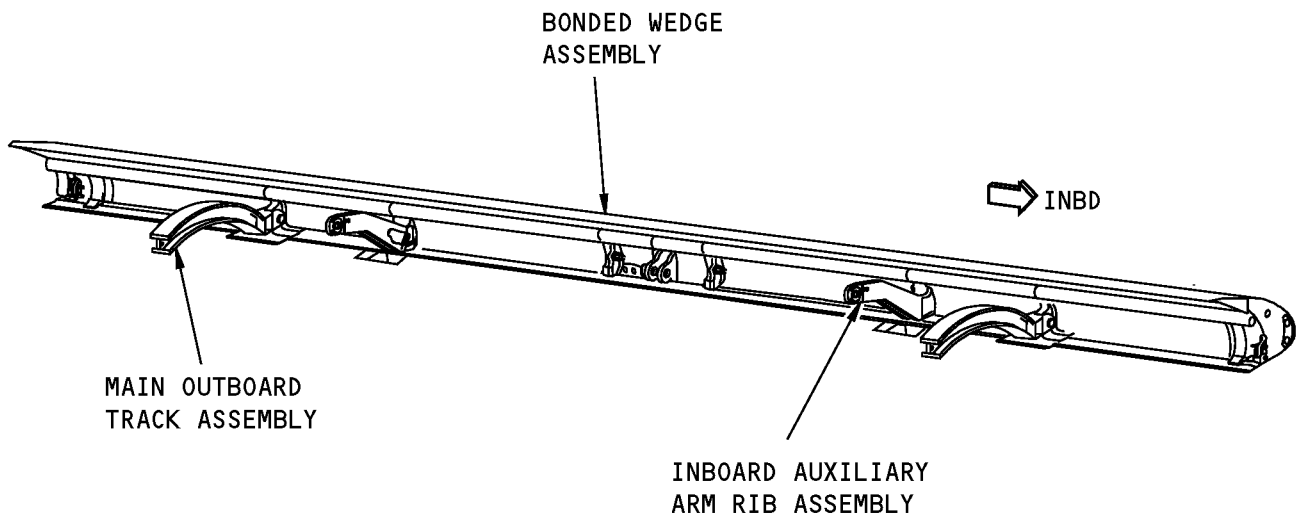
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DESCRIPTION AND OPERATION

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Wing Leading Edge No. 1 and 8 Slat Assemblies
Figure 1

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DESCRIPTION AND OPERATION

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TESTING AND FAULT ISOLATION

(NOT APPLICABLE)

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TESTING AND FAULT ISOLATION

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DISASSEMBLY

1. General

- A. This procedure has the data necessary to disassemble the wing leading edge No. 1 and 8 slat assemblies.
- B. Disassemble this component sufficiently to isolate the defects, do the necessary repairs, and put the component back to a serviceable condition.

2. Disassembly

- A. Procedure
 - (1) Use standard industry procedures to disassemble this component.

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DISASSEMBLY

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CLEANING

1. General

- A. This procedure has the data necessary to clean the wing leading edge No. 1 and 8 slat assemblies.
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

2. cleaning

A. References

Reference	Title
SOPM 20-30-01	CLEANING AND RELUBRICATING BEARINGS
SOPM 20-30-03	GENERAL CLEANING PROCEDURES

B. Procedure

CAUTION: DO NOT VAPOR DEGREASE EPOXY BONDED STRUCTURES WITH CHLORINATED CLEANING AGENTS SUCH AS METHYLENE CHLORIDE, TRICHLOROETHYLENE, AND TRICHLOROETHANE. CHLORINATED CLEANING AGENTS WILL CAUSE DAMAGE TO EPOXY BONDED STRUCTURES. 1,1,1-TRICHLOROETHANE IS ONE OF THE SOLVENTS ALLOWED FOR CLEANING COMPOSITE COMPONENTS. DO NOT SUBMERGE PARTS IN THE SOLVENT OR ALLOW STANDING SOLVENT ON THE PARTS OR DAMAGE MAY OCCUR. USE 1,1, 1-TRICHLOROETHANE ONLY AS A WIPE SOLVENT.

CAUTION: VAPOR DEGREASING OF TITANIUM PARTS IS RESTRICTED. SEE SOPM 20-30-03 FOR INSTRUCTION.

- (1) Clean the bearings (IPL Figure 1; 508) (IPL Figure 4; 520) as specified in SOPM 20-30-01 .
- (2) Use standard industry procedures and refer to SOPM 20-30-03 to clean all parts.

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CLEANING

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CHECK

1. General

- A. This procedure has the data necessary to find defects in the material of the specified parts.
- B. Refer to FITS AND CLEARANCES for the design dimension and wear limits.
- C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

2. Check

A. References

Reference	Title
SOPM 20-20-01	MAGNETIC PARTICLE INSPECTION
SOPM 20-20-02	PENETRANT METHODS OF INSPECTION
737 NDT 51-00-01	Non-Destructive Testing
737 NDT 51-00-02	Non-Destructive Testing
737 NDT 51-05-01	Non-Destructive Testing

B. Procedure

- (1) Use standard industry procedures to do a visual check of all the parts for defects. Do the penetrant or the magnetic particle check if the visual check shows possible damage or if you suspect possible damage on the parts listed below:
- (2) Do a magnetic particle check (SOPM 20-20-01) of these parts:
 - (a) Track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44)
 - (b) Fitting (IPL Figure 1; 136, 572) (IPL Figure 4; 152, 592)
- (3) Do a penetrant check (SOPM 20-20-02) of these parts:
 - (a) Rib (IPL Figure 1; 228, 232, 236, 240, 512, 516, 520, 524, 688, 692) (IPL Figure 4; 244A, 248A, 252, 256, 257, 258, 524, 528, 532, 536)
 - (b) Guard (IPL Figure 2; 30)
 - (c) Rib (IPL Figure 2; 35, 40, 45, 50)
 - (d) Rib (IPL Figure 3; 30, 35, 40, 45)
- (4) Do a check for cracks, corrosion, or missing potting and sealant at the ends of the panel.
- (5) Do a check of the honeycomb structure and bonded parts for evidence of delamination, internal water, scratches, and contour defects:
 - (a) If you see delamination or impact damage when you do a visual check, do an ultrasonic inspection or a tap test to find all damage. Refer to the 737 Nondestructive Test Manual, Part 4, 737 NDT 51-00-02 of Part 1, 737 NDT 51-05-01.

NOTE: For the tap test, use a solid metal disk and tap the surface area lightly but firmly. The void areas will produce a dull sound as opposed to a sharp sound on a solid bonded area. Refer to the 737 Nondestructive Test Manual, Part 1, 737 NDT 51-05-01.

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CHECK

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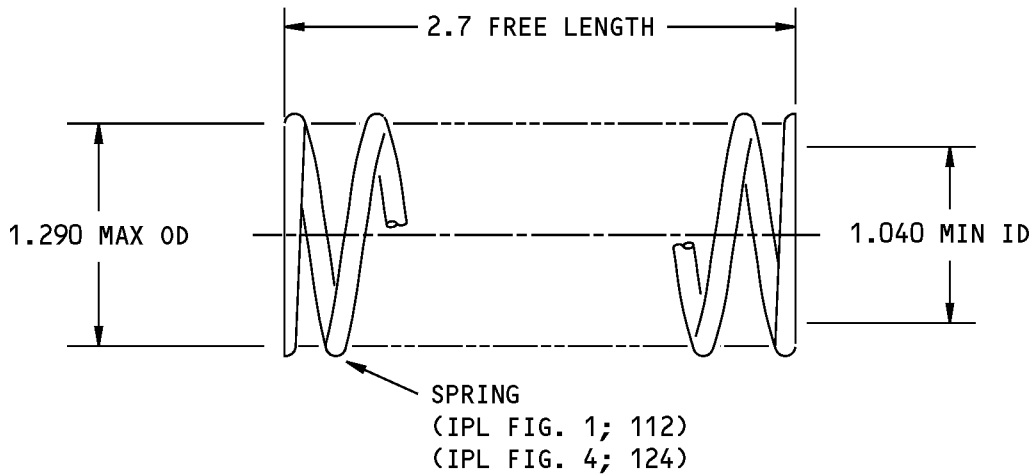
- (b) Do a check on areas you suspect of containing water with the radiographic or thermographic method. Refer to the 737 Nondestructive Test Manual, Part 2, 737 NDT 51-00-01 or Part 9, 737 NDT 51-00-01.
- (c) Do a check on the edges of the panel carefully for cuts and abrasions. Delamination starts very easily from damage to an edgemember of honeycomb panel.
- (6) Refer to the applicable 737 Structural Repair Manuals section for allowable damage and repair data.
- (7) Do a check on the spring (IPL Figure 1; 112) (IPL Figure 4; 124) as specified in CHECK, Figure 501 .

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NOTE:

MAXIMUM CHECK LOAD: 20.5-24.5 POUNDS AT 0.95 INCHES

MINIMUM CHECK LOAD: 15.1-19.1 POUNDS AT 1.37 INCHES

MAXIMUM SOLID HEIGHT: 0.75 INCHES

ALL DIMENSIONS ARE IN INCHES

114W4709-1 Spring Check
Figure 501

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CHECK
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REPAIR

1. General

- A. Instructions for repair, refinish, and replacement of the specified subassembly parts are included in each REPAIR when applicable:

Table 601:

PART NUMBER	NAME	REPAIR
—	REFINISH OF OTHER PARTS	1-1
114A7011	OUTBOARD/INBOARD MAIN	2-1, 2-2
114A7012	TRACK RIB ASSEMBLY	
114A7111	OUTBOARD/INBOARD	3-1, 3-2
114A7112	AUXILIARY RIB ASSEMBLY	
114A7211	ACTUATOR RIB ASSEMBLY	4-1, 4-2
114A7311	OUTBOARD/INBOARD UP STOP	5-1, 5-2
114A7312	RIB ASSEMBLY	
114A7411	OUTBOARD/INBOARD END	6-1, 6-2
114A7412	RIB ASSEMBLY	
114A7511	OUTBOARD/INBOARD MAIN	7-1, 7-2
114A7512	TRACK ASSEMBLY	

2. Dimensioning Symbols

- A. Standard True Position Dimensioning Symbols used in the applicable repair procedures are shown in REPAIR-GENERAL, Figure 601.

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REPAIR - GENERAL

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- STRAIGHTNESS
- ▭ FLATNESS
- ⊥ PERPENDICULARITY (OR SQUARENESS)
- // PARALLELISM
- ROUNDNESS
- ⊘ CYLINDRICITY
- ⌒ PROFILE OF A LINE
- ⌓ PROFILE OF A SURFACE
- ◎ CONCENTRICITY
- ≡ SYMMETRY
- ∠ ANGULARITY
- ↗ RUNOUT
- ↗↗ TOTAL RUNOUT
- ⊔ COUNTERBORE OR SPOTFACE
- ∇ COUNTERSINK
- ⊕ THEORETICAL EXACT POSITION OF A FEATURE (TRUE POSITION)
- ∅ DIAMETER
- S ∅ SPHERICAL DIAMETER
- R RADIUS
- SR SPHERICAL RADIUS
- () REFERENCE
- BASIC A THEORETICALLY EXACT DIMENSION USED TO DESCRIBE SIZE, SHAPE OR LOCATION OF A FEATURE. FROM THIS FEATURE PERMISSIBLE VARIATIONS ARE ESTABLISHED BY TOLERANCES ON OTHER DIMENSIONS OR NOTES.
- (DIM) DIM
- A- DATUM
- Ⓜ MAXIMUM MATERIAL CONDITION (MMC)
- Ⓛ LEAST MATERIAL CONDITION (LMC)
- Ⓢ REGARDLESS OF FEATURE SIZE (RFS)
- Ⓟ PROJECTED TOLERANCE ZONE
- FIM FULL INDICATOR MOVEMENT

EXAMPLES

- 0.002 STRAIGHT WITHIN 0.002
- ⊥ 0.002 B PERPENDICULAR TO DATUM B WITHIN 0.002
- // 0.002 A PARALLEL TO DATUM A WITHIN 0.002
- 0.002 ROUND WITHIN 0.002
- ⊘ 0.010 CYLINDRICAL SURFACE MUST LIE BETWEEN TWO CONCENTRIC CYLINDERS, ONE OF WHICH HAS A RADIUS 0.010 INCH GREATER THAN THE OTHER
- ⌒ 0.006 A EACH LINE ELEMENT OF THE SURFACE AT ANY CROSS SECTION MUST LIE BETWEEN TWO PROFILE BOUNDARIES 0.006 INCH APART RELATIVE TO DATUM A
- ⌓ 0.020 A SURFACES MUST LIE WITHIN PARALLEL BOUNDARIES 0.020 INCH APART AND EQUALLY DISPOSED ABOUT TRUE PROFILE
- ◎ ∅ 0.0005 C CONCENTRIC TO DATUM C WITHIN 0.0005 DIAMETER
- ≡ 0.010 A SYMMETRICAL WITH DATUM A WITHIN 0.010
- ∠ 0.005 A ANGULAR TOLERANCE 0.005 WITH DATUM A
- ⊕ ∅ 0.002 Ⓢ B LOCATED AT TRUE POSITION WITHIN 0.002 DIA RELATIVE TO DATUM B, REGARDLESS OF FEATURE SIZE
- ⊥ ∅ 0.010 Ⓜ A AXIS IS TOTALLY WITHIN A CYLINDER OF 0.010 INCH DIAMETER, PERPENDICULAR TO DATUM A, AND EXTENDING 0.510 INCH ABOVE DATUM A, MAXIMUM MATERIAL CONDITION
- 0.510 Ⓟ
- 2.000 THEORETICALLY EXACT DIMENSION IS 2.000
- OR
- 2.000 BSC

True Position Dimensioning Symbols
Figure 601

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REPAIR - GENERAL

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REFINISH OF OTHER PARTS - REPAIR 1-1

1. General

- A. This procedure has the data necessary to refinish the parts which are not given in the specified repairs.
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to IPL Figure 1, IPL Figure 2 and IPL Figure 4 for item numbers.

2. Refinish of other parts

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00033	Coating - Exterior Protective Enamel, Flexibility Use	BMS10-60, Type II
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I
C00766	Primer - Nonchromated (For Non-Metalic Composites)	BMS10-103, Type 1
C00767	Coating - Anti-Static Coating	BMS10-21, Type III

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02.

- (1) Instructions for the repair of the parts listed in REPAIR 1-1, Table 601 is for repair of the initial finish.

Table 601: Refinish Details

IPL FIG. & ITEM	MATERIAL	FINISH
IPL Figure 1		
Slat assembly (1A, 4)		See REPAIR 1-1, Figure 601.

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Table 601: Refinish Details (Continued)

IPL FIG. & ITEM	MATERIAL	FINISH
Retainer (24A, 596A, 600A)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Tab (84, 88)	Epoxy impregnated glass woven fabric	On tool side: Prepare surface (SRF-14.672). Prepare surface and apply coating, C00767 (F-14.685). Prepare surface and apply primer, C00766 (F-14.692). Apply enamel coating, C00033 (F-19.39-707). On bag side: Prepare surface and apply primer, C00766 (F-14.692). Apply enamel coating, C00033 (F-19.39-707).
Washer (100, 102)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Sleeve (104)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Retainer (108)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Fitting (136, 572)	15-5PH CRES 180-200 ksi	Cadmium plate (F-16.06). Apply primer, C00175 (F-19.47) all over except in holes.
Retainer (288, 292, 740A, 744, 748, 752, 756)	Aluminum alloy	Chemical treat (F-17.07). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Bracket (344, 348, 352, 356)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47).
Target (364)	HYMU80 annealed	Cadmium plate (F-15.06). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Tab (388, 392, 396, 400)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707). No enamel in holes.
Support (404, 408, 412, 416)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033(F-19.39-707).
Wedge assembly (828, 832)		Manually apply coating (F-17.10). Apply primer, C00175 (F-19.47) all over.
Beam (836, 840)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00259 (F-20.02).
Vortilion (843, 844)	Aluminum alloy	Abrasive blast prior to (F-17.29) Boric acid - sulfuric acid anodize.
IPL Figure 2		
Guard (30)	Copper Beryllium	Cadmium plate (F-15.06). Apply primer, C00259 (F-20.02).
IPL Figure 4		
Slat assembly (1A, 4)		See REPAIR 1-1, Figure 601.

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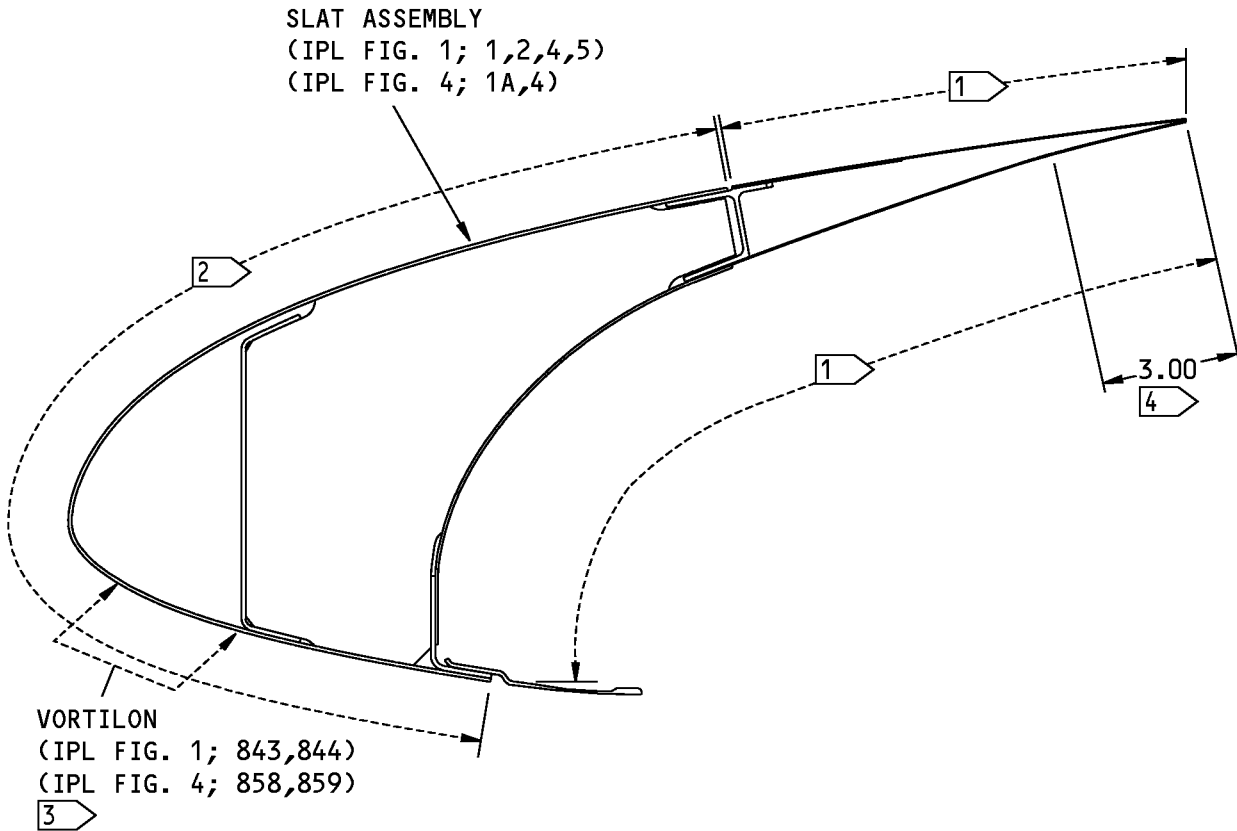
Table 601: Refinish Details (Continued)

IPL FIG. & ITEM	MATERIAL	FINISH
Retainer (76, 616, 620)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Tab (92, 96)	Epoxy impregnated glass woven fabric	On tool side: Prepare surface (SRF-14.672). Prepare surface and apply coating, C00767 (F-14.685). Prepare surface and apply primer, C00766(F-14.692). Apply enamel coating, C00033 (F-19.39-707). On bag side: Prepare surface and apply primer, C00766 (F-14.692). Apply enamel coating, C00033 (F-19.39-707).
Washer (108, 112)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Sleeve (116)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Sleeve (116A)	6013-T6 aluminum alloy (optional to 2024-T42)	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) and apply primer, C00259 (F-20.02).
	2024-T42 aluminum alloy (optional to 6013-T6)	Chemical treat (F-17.07) and boric acid - sulfuric acid anodize (F-17.31) primer, C00259 (F-20.02).
Retainer (120)	Aluminum alloy	Chemical treat and apply primer, C00259 (F-18.06).
Fitting (152, 592)	15-5PH CRES 180-200 ksi	Cadmium plate (F-16.06). Apply primer, C00175 (F-19.47) all over except in holes.
Retainer (304, 308, 756, 760, 764, 768, 772)	Aluminum alloy	Chemical treat (F-17.07). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Bracket (360, 364, 368, 372)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47).
Target (380)	HYMU80 annealed	Cadmium plate (F-15.06). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Tab (404, 408, 412, 416)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707). No enamel in holes.
Support (420, 424, 428, 432)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Wedge assembly (844, 848)		Manually apply coating (F-17.10). Apply primer, C00175 (F-19.47) all over.
Beam (852, 856)	Aluminum alloy	Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31). Apply primer, C00259 (F-20.02).
Beam (885, 888, 891, 894)	Aluminum alloy	Boric acid - sulfuric acid anodize (F-17.41). Apply primer, C00259 (F-20.02).

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- 1 APPLY BMS 10-60, TYPE 2 GLOSS ENAMEL (F-19.39-707) TO THE UPPER SURFACE OF WEDGE, COVE AREA, AND THE OUTER SURFACES OF THE FLAT ASSEMBLY
- 2 NO FINISH (F-25.01) ON LEADING EDGE SKIN
- 3 ONLY APPLIES TO 114A5010-3,-4,-9,-10,-11,-12,-201,-202
- 4 APPLY BMS 10-86, TYPE 1 OR TYPE 2 TEFLON (F-14.9625) TO A THICKNESS OF 0.005-0.010 ON THE T.E. WEDGE ASSEMBLY LOWER SURFACE

ALL DIMENSIONS ARE IN INCHES

114A5010 Slat Assembly Refinish
Figure 601

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OUTBOARD/INBOARD MAIN TRACK RIB ASSEMBLY - REPAIR 2-1

114A7011-1, -2, 114A7012-1, -2, -201, -202

1. General

- A. This procedure has the data necessary to repair and refinish the outboard/inboard main track rib assembly (IPL Figure 1; 492, 496, 500, 504) (IPL Figure 4; 504, 508, 512, 516).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL (57-43-01/601, REPAIR-GENERAL) for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.

2. Bearing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bearing (IPL Figure 1; 508) (IPL Figure 4; 520) from the rib (IPL Figure 1; 512, 520) (IPL Figure 4; 524, 532) (SOPM 20-50-03).
- (2) Install the bearing (IPL Figure 1; 508) (IPL Figure 4; 520) on the rib (IPL Figure 1; 512, 520) (IPL Figure 4; 524, 532) with sealant, A00247 and roller swage (SOPM 20-50-03).

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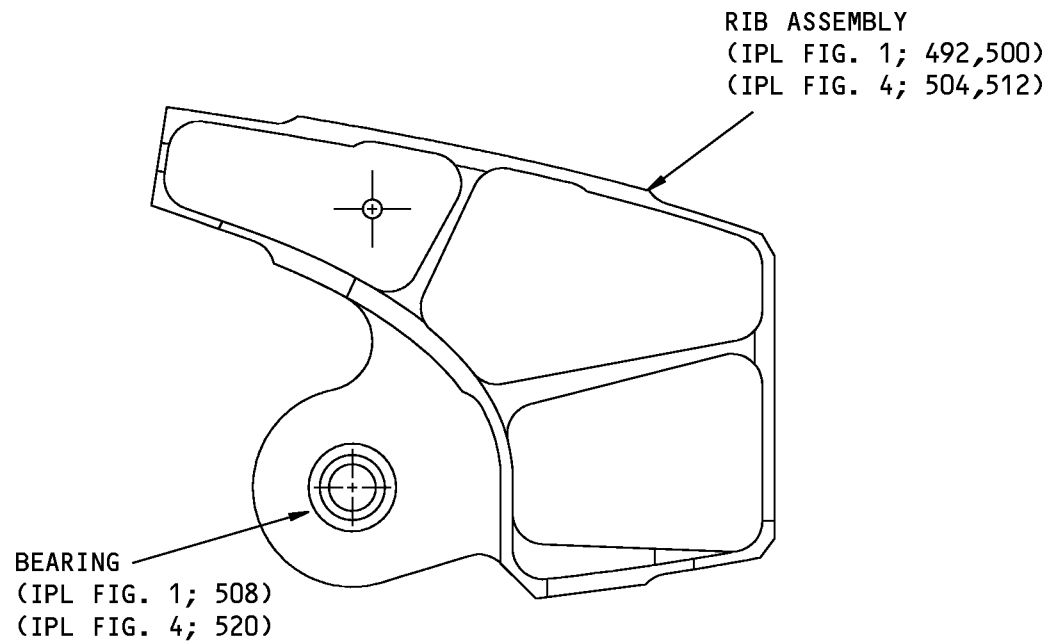
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114A7011-1; 114A7012-1,-201 SHOWN
114A7011-2; 114A7012-2,-202 OPPOSITE

125/ ALL MACHINED SURFACES UNLESS
SHOWN DIFFERENTLY

ALL DIMENSIONS ARE IN INCHES

114A7011-1, -2; 114A7012-1, -2, -201, -202 Inboard and Outboard Main Track Rib Assemblies Repair
Figure 601

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REPAIR 2-1

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RIB - REPAIR 2-2

114A7011-3, -4, 114A7012-3, -4, -203, -204

1. General

- A. This procedure has the data necessary to refinish the rib (IPL Figure 1; 512, 516, 520, 524) (IPL Figure 4; 524, 528, 532, 532A, 536, 536A).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL (57-43-01/601, REPAIR-GENERAL) for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.
- E. General repair details:
 - (1) Materials: Aluminum alloy
 - (2) Shot peen: All repaired surfaces, except in holes
Intensity 0.004A-0.007A
Coverage 1.0 (automated), 2.0 (manual)

2. Rib Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

- C. Procedure (REPAIR 2-2, Figure 601).

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02.

- (1) (For all except P/N 114A7012-203, -204) Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) all over the rib (IPL Figure 1; 512, 520) (IPL Figure 4; 524, 532).
- (2) (For P/N 114A7012-203, -204) Boric acid - sulfuric acid anodize (F-17.41) all over the rib (IPL Figure 4; 532A, 536A).
- (3) Apply primer, C00175 (F-19.47) to the rib (IPL Figure 1; 512, 520) (IPL Figure 4; 524, 532). No primer in the bearing hole.

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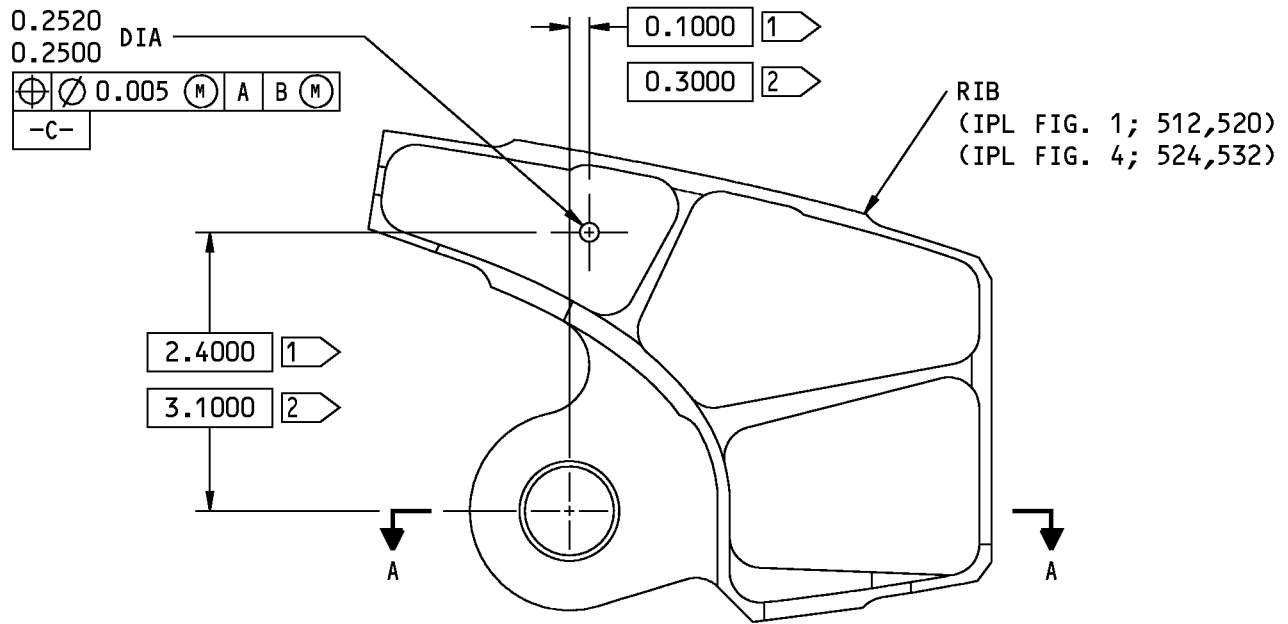
REPAIR 2-2

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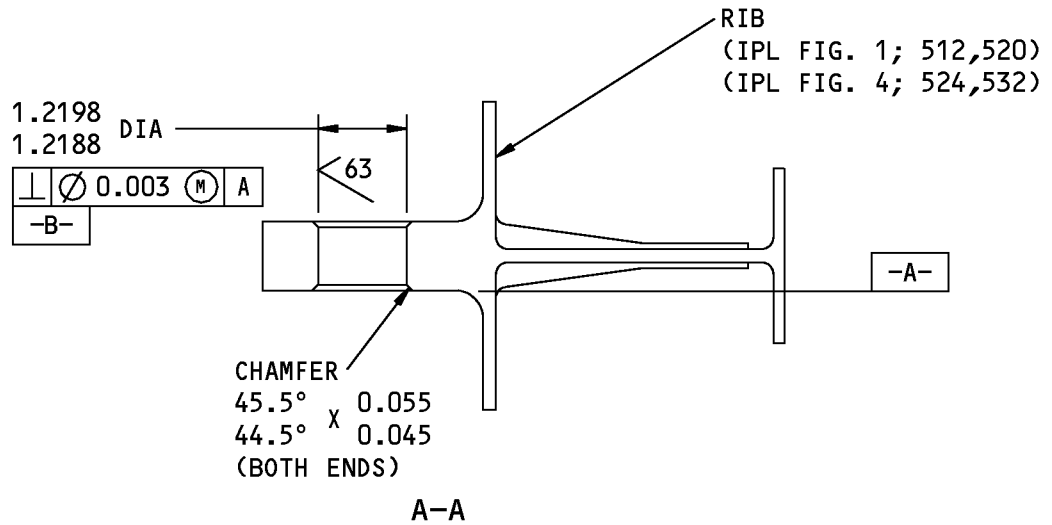
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114A7011-3; 114W7012-3,-203 SHOWN
114A7011-4; 114W7012-4,-204 OPPOSITE



- 1 FOR 114A7011-3,-4
- 2 FOR 114A7012-3,-4,-203,-204

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ALL DIMENSIONS ARE IN INCHES

114A7011-3, -4; 114A7012-3, -4, -203, -204 Inboard and Outboard Main Track Rib Assemblies Repair
Figure 601

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REPAIR 2-2
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OUTBOARD/INBOARD AUXILIARY ARM RIB ASSEMBLY - REPAIR 3-1

114A7111-1, -2, 114A7112-1, -2

1. General

- A. This procedure has the data necessary to repair and refinish the outboard/inboard auxiliary arm rib assembly (1A, 5, 10, 15).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL (57-43-01/601, REPAIR-GENERAL) for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 2 for item numbers.

2. Bushing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

WARNING: THE SIDE GUARDS (30) ARE MADE OF COPPER BERYLLIUM. REFER TO SOPM 20-10-09 FOR HANDLING PROCEDURE.

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

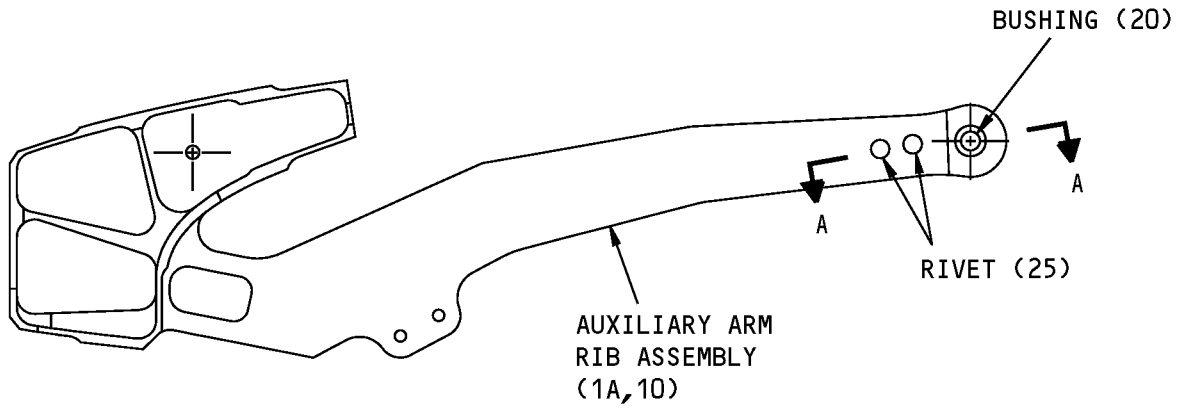
- (1) Remove the rivets (25) to remove the side guards (30) from the rib (35, 45).
- (2) Remove the bushings (20) from the rib (35, 45) (SOPM 20-50-03).
- (3) Install the bushings (20) on the rib (35, 45) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03) .
- (4) Ream the inside diameter of bushings (20) to the dimensions shown on REPAIR 3-1, Figure 601.
- (5) Break all sharp edges (0.0200 inch minimum radius).
- (6) Install the side guards (30) on the rib (35, 45) with rivets (25).

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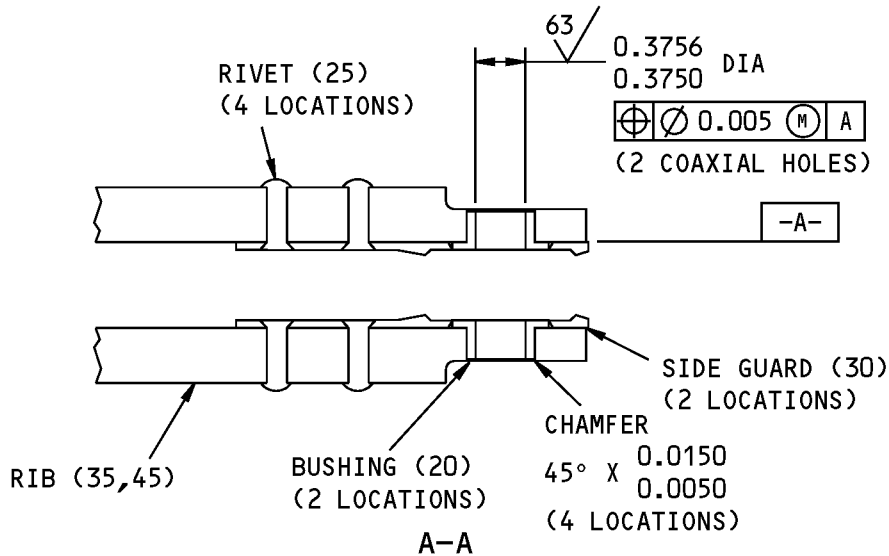
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114A7111-1, 114A7112-1 SHOWN
 114A7111-2, 114A7112-2 OPPOSITE



NOTE: SEE REPAIR 3-2 FOR BUSHING HOLE LOCATION

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ITEM NUMBERS REFER TO IPL FIG. 2
 ALL DIMENSIONS ARE IN INCHES

114A7111-1,-2 114A7112-1,-2 Inboard and Outboard Auxiliary Rib Assembly Repair
 Figure 601

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REPAIR 3-1
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RIB - REPAIR 3-2

114A7111-3, -4, 114A7112-3, -4

1. General

- A. This procedure has the data necessary to refinish the rib (35, 40, 45, 50).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL (57-43-01/601, REPAIR-GENERAL) for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 2 for item numbers.
- E. General repair details:
 - (1) Materials: Aluminum alloy
 - (2) Shot peen: All repaired surfaces, except in holes
Intensity 0.004A-0.007A
Coverage 1.0 (automated), 2.0 (manual)
No intensity check required on the facing area in the clevis slot. Manual peening is allowed, check for coverage 1.0 only.

2. Rib Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00319	Primer - Urethane Compatible, Corrosion Resistant	BMS10-79, Type II

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure (REPAIR 3-2, Figure 601)

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) all over the rib (35, 45).
- (2) Apply primer, C00319 (F-19.47) to the rib (35, 45). No primer in the bearing hole.

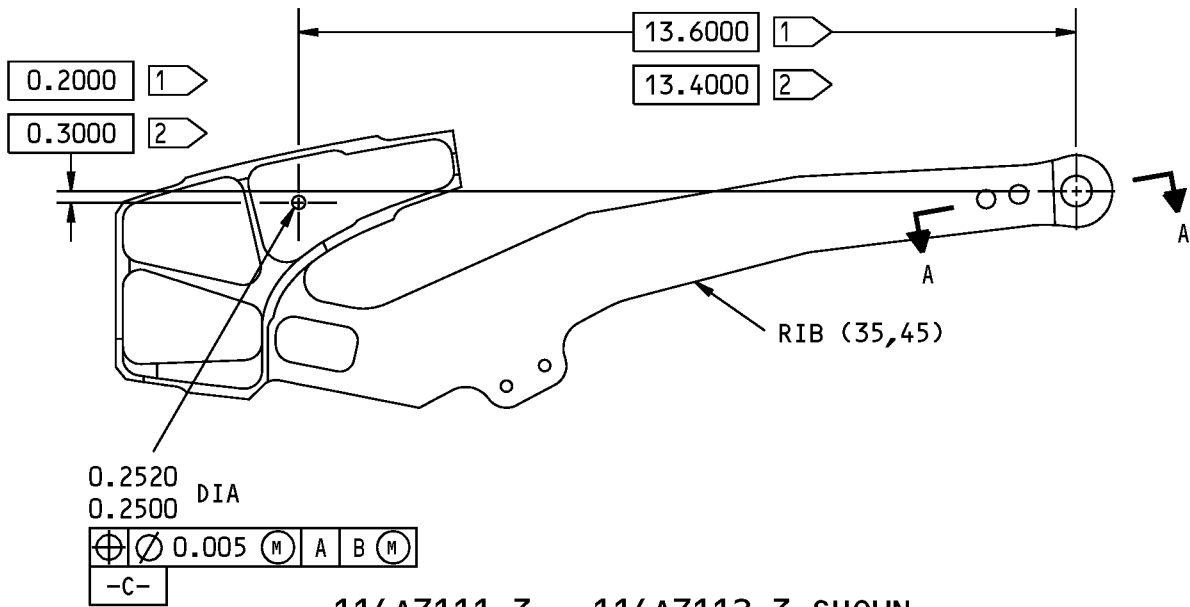
57-43-01

REPAIR 3-2

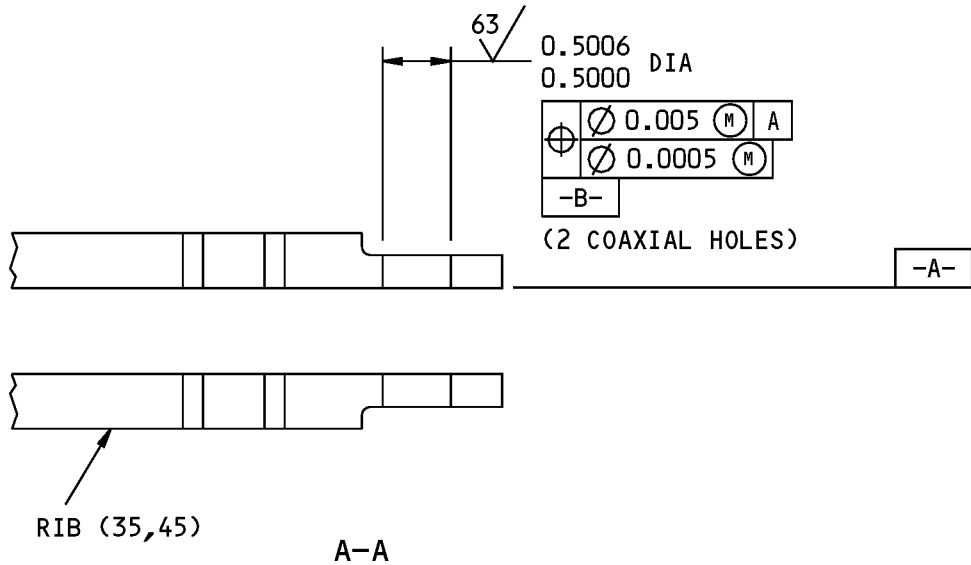
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114A7111-3, -114A7112-3 SHOWN
114A7111-4, -114A7112-4 OPPOSITE



- 1 FOR 114A7111-3,-4
- 2 FOR 114A7112-3,-4

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ITEM NUMBERS REFER TO IPL FIG. 2
ALL DIMENSIONS ARE IN INCHES

114A7111-3,-4 114A7112-3,-4 Rib Repair
Figure 601

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REPAIR 3-2
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ACTUATOR ARM RIB ASSEMBLY - REPAIR 4-1

114A7211-1, -2, -201, -202

1. General

- A. This procedure has the data necessary to repair and refinish the actuator rib assembly (IPL Figure 1; 672, 676) (IPL Figure 4; 688, 692).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.

2. Bushing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

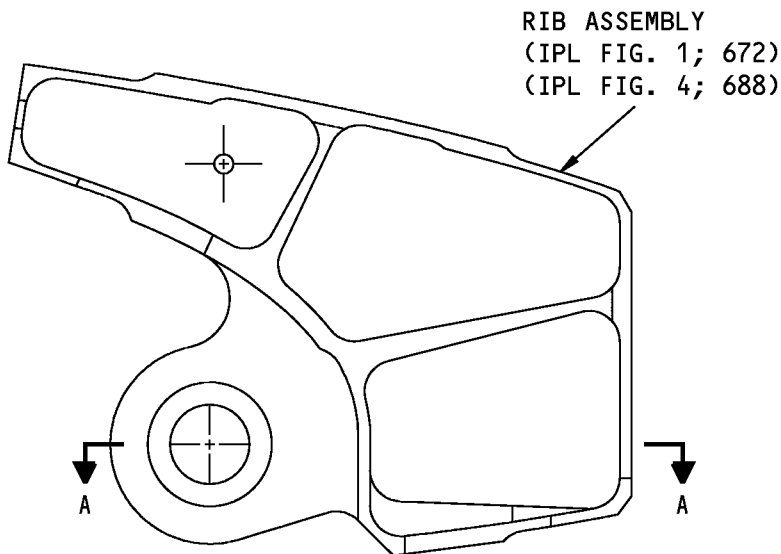
- (1) Remove the bushings (IPL Figure 1; 680, 684) (IPL Figure 4; 696, 700) from the rib (IPL Figure 1; 688) (IPL Figure 4; 704).
- (2) Install the bushings (IPL Figure 1; 680, 684) (IPL Figure 4; 696, 700) on the rib (IPL Figure 1; 688) (IPL Figure 4; 704) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03).
- (3) Ream the inside diameter of bushings (IPL Figure 1; 680, 684) (IPL Figure 4; 696, 700) to the dimensions shown on REPAIR 4-1, Figure 601.
- (4) Break all sharp edges (0.0200 inch minimum radius).

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REPAIR 4-1
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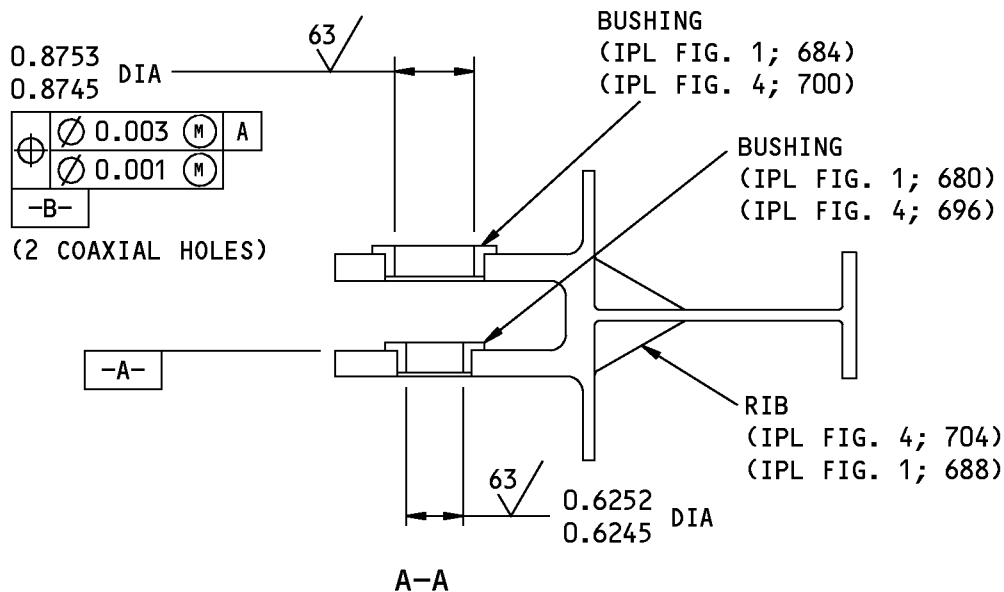


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RIB ASSEMBLY
(IPL FIG. 1; 672)
(IPL FIG. 4; 688)

114A7211-1,-201 SHOWN
114A7211-2,-202 OPPOSITE



NOTE: SEE REPAIR 4-2 FOR BUSHING HOLE LOCATION

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGES

ALL DIMENSIONS ARE IN INCHES

114A7211-1, -2, -201, -202 Actuator Rib Assembly Repair
Figure 601

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REPAIR 4-1
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RIB - REPAIR 4-2

114A7211-3, -4, -203, -204

1. General

- A. This procedure has the data necessary to refinish the rib (IPL Figure 1; 688, 692) (IPL Figure 4; 704, 704A, 708, 708A).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.
- E. General repair details:
 - (1) Materials: Aluminum alloy
 - (2) Shot peen: All repaired surfaces, including bores
Intensity 0.004A-0.007A
Coverage 1.0 (automated), 2.0 (manual)

2. Rib Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

- C. Procedure (REPAIR 4-2, Figure 601)

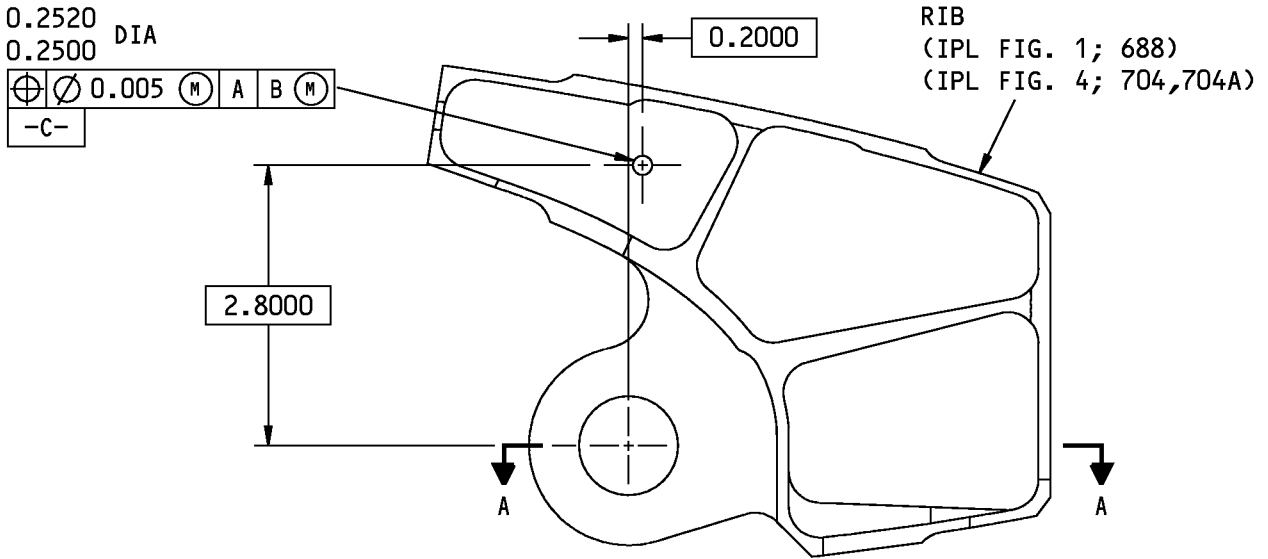
NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02.

- (1) (For all except P/N 114A7211-203, -204) Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) all over the rib (IPL Figure 1; 688) (IPL Figure 4; 704).
- (2) (For P/N 114A7211-203, -204) Boric acid - sulfuric acid anodize (F-17.41) all over the rib (IPL Figure 4; 704A).
- (3) Apply primer, C00175 (F-19.47) to the rib IPL Figure 1; 688) (IPL Figure 4; 704). No primer in the bearing hole.

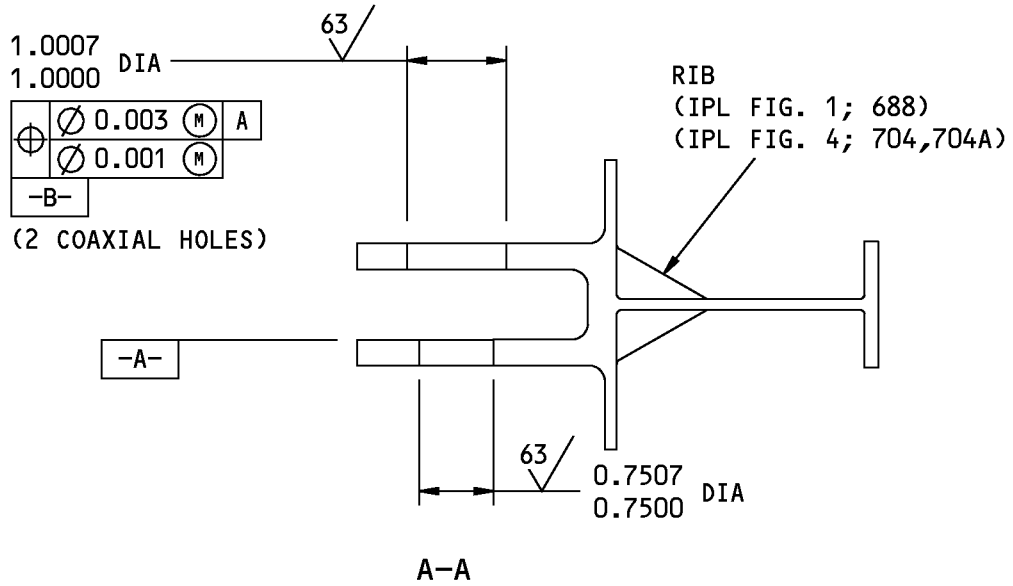
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REPAIR 4-2
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114A7211-3,-203 SHOWN
114A7211-4,-204 OPPOSITE



125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ALL DIMENSIONS ARE IN INCHES

114A7211-3, -4, -203, -204 Actuator Rib Assembly Repair
Figure 601

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REPAIR 4-2

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OUTBOARD/INBOARD UP STOP RIB ASSEMBLY - REPAIR 5-1

114A7311-1, -2, -201, -202, 114A7312-1, -2, -201, -202

1. General

- A. This procedure has the data necessary to repair and refinish the outboard/inboard up stop rib assembly (1A, 5, 10, 15).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 3 for item numbers.

2. Bushing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

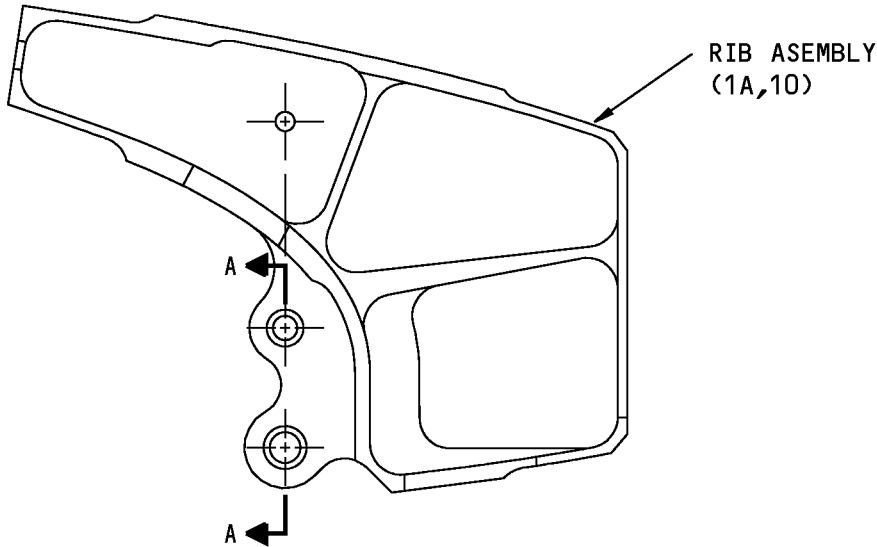
NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bushings (20, 25) from the rib (30, 40).
- (2) Install the bushings (20, 25) on the rib (30, 40) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03).
- (3) Ream the inside diameter of bushings (20, 25) to the dimensions shown on REPAIR 5-1, Figure 601.
- (4) Break all sharp edges (0.02-0.023 inch radius). Visible inspection acceptable.

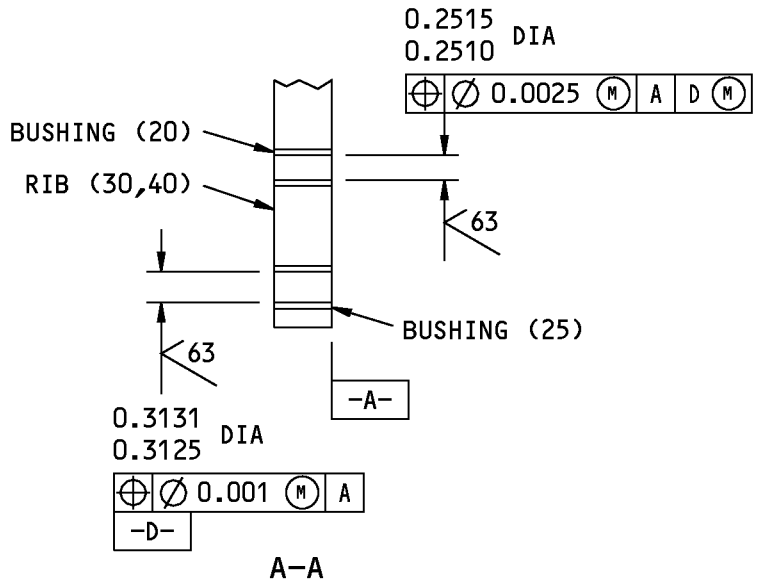
57-43-01

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114A7311-1,-201; 114W7312-1,-201 SHOWN
 114A7311-2,-202; 114W7312-2,-202 OPPOSITE



NOTE: SEE REPAIR 5-2 FOR BUSHING HOLE LOCATION

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ITEM NUMBERS REFER TO IPL FIG. 3
 ALL DIMENSIONS ARE IN INCHES

114A7311-1, -2, -201, -202; 114A7312-1, -2, -201, -202 Upstop Inboard and Outboard Rib Assembly
 Repair
 Figure 601

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REPAIR 5-1
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RIB - REPAIR 5-2

114A7311-3, -4, -203, -204, 114A7312-3, -4, -203, -204

1. General

- A. This procedure has the data necessary to refinish the rib (30, 30A, 35, 35A, 40, 40A, 45, 45A).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 3 for item numbers.
- E. General repair details:
 - (1) Materials: Aluminum alloy
 - (2) Shot peen: All repaired surfaces, except in holes
Intensity 0.004A-0.007A
Coverage 1.0 (automated), 2.0 (manual)

2. Rib Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

- C. Procedure (REPAIR 5-2, Figure 601)

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02.

- (1) (For all except P/N 114A7311-203, -204; P/N 114A7312-203, -204) Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) all over the rib (30, 40).
- (2) (For P/N 114A7311-203, -204; P/N 114A7312-203, -204) Boric acid - sulfuric acid anodize (F-17.41) all over the rib (30A, 40A).
- (3) Apply one coat of primer, C00175 (F-19.47) to the rib (30, 40). No primer in the bearing hole.

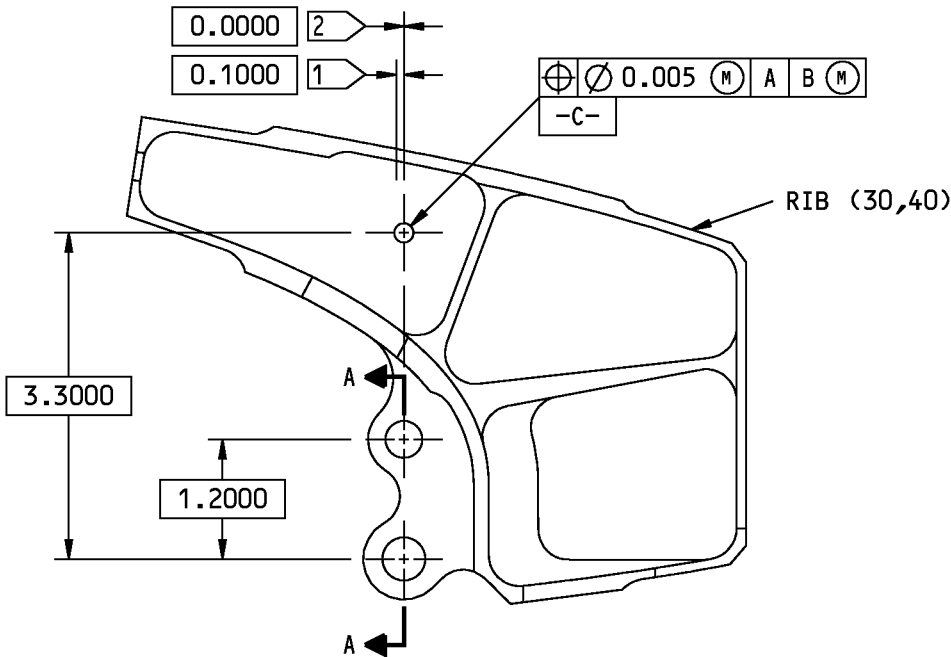
57-43-01

REPAIR 5-2

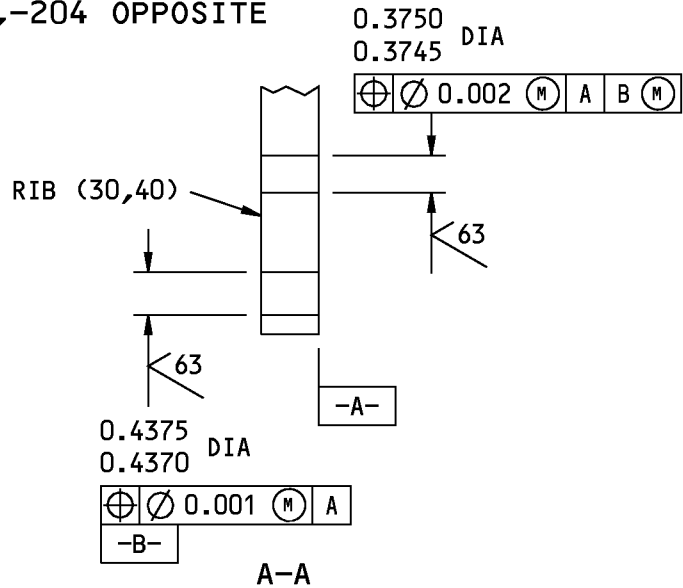
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114A7311-3,-203; 114A7312-3,-203 SHOWN
 114A7311-4,-204; 114A7312-4,-204 OPPOSITE



- 1 FOR 114A7311
- 2 FOR 114A7312

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ITEM NUMBERS REFER TO IPL FIG. 3
 ALL DIMENSIONS ARE IN INCHES

114A7311-3, -4, -203, -204; 114A7312-3, -4, -203, -204 Rib Repair
 Figure 601

57-43-01



COMPONENT MAINTENANCE MANUAL

OUTBOARD/INBOARD END RIB ASSEMBLY - REPAIR 6-1

114A7411-1, -2, -5, -6, 114A7412-1, -2, -201, -202

1. General

- A. This procedure has the data necessary to repair and refinish the outboard/inboard end arm rib assembly (IPL Figure 1; 200, 204, 208, 212) (IPL Figure 4; 216A, 220A, 224, 228, 229, 230).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.

2. Bushing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

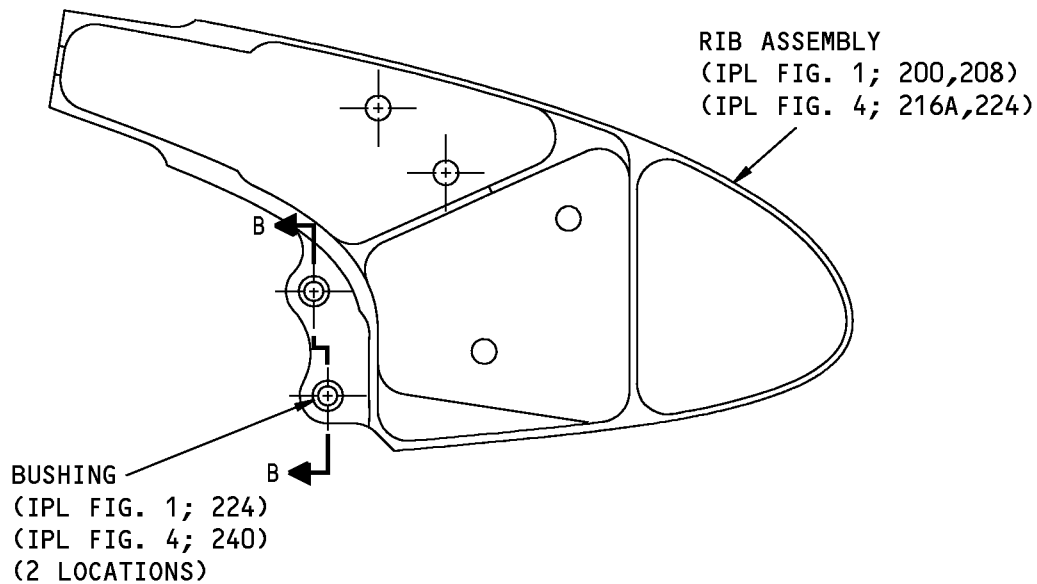
NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bushings (IPL Figure 1; 224) (IPL Figure 4; 240) from the rib (IPL Figure 1; 228, 236) (IPL Figure 4; 244A, 252).
- (2) Install the bushings (IPL Figure 1; 224) (IPL Figure 4; 240) on the rib (IPL Figure 1; 228, 236) (IPL Figure 4; 244A, 252) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03).
- (3) Ream the inside diameter of bushings (IPL Figure 1; 224) (IPL Figure 4; 240) to the dimensions shown on REPAIR 6-1, Figure 601.
- (4) Break all sharp edges.

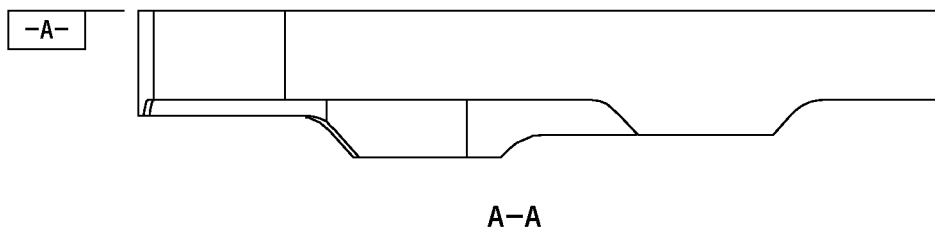
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114A7411-1,-5; 114A7412-1,-201 SHOWN
 114A7411-2,-6; 114A7412-2,-202 OPPOSITE

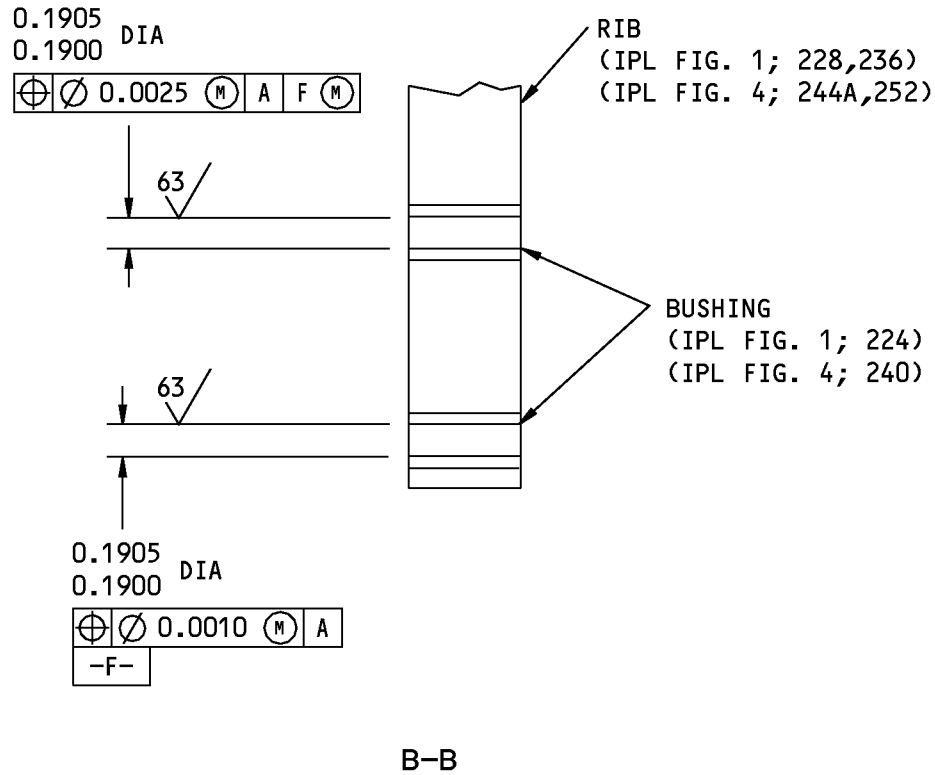


114A7411-1, -2, -5, -6; 114A7412-1, -2, -201, -202 Inboard and Outboard End Rib Assembly Repair
 Figure 601 (Sheet 1 of 2)

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REPAIR 6-1
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NOTE: SEE REPAIR 6-2 FOR BUSHING HOLE LOCATION

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ALL DIMENSIONS ARE IN INCHES

114A7411-1, -2, -5, -6; 114A7412-1, -2, -201, -202 Inboard and Outboard End Rib Assembly Repair
Figure 601 (Sheet 2 of 2)

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REPAIR 6-1
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COMPONENT MAINTENANCE MANUAL

RIB - REPAIR 6-2

114A7411-3, -4, -7, -8, 114A7412-3, -4, -203, -204

1. General

- A. This procedure has the data necessary to refinish the rib (IPL Figure 1; 228, 232, 236, 240) (IPL Figure 4; 244A, 248A, 252, 256, 257, 258).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.
- E. General repair details:
 - (1) Material: Aluminum alloy

2. Rib Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

- C. Procedure (REPAIR 6-2, Figure 601)

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02.

- (1) (For all except P/N 114A7412-203, -204) Boric acid - sulfuric acid anodize or chromic acid anodize (F-17.31) all over the rib (IPL Figure 1; 228, 236) (IPL Figure 4; 244A, 252).
- (2) (For P/N 114A7412-203, -204) Boric acid - sulfuric acid anodize (F-17.41) all over the rib (IPL Figure 4; 257)
- (3) Apply primer, C00175 (F-19.47) to the rib (IPL Figure 1; 228, 236) (IPL Figure 4; 244A, 252). No primer in the bushing hole.

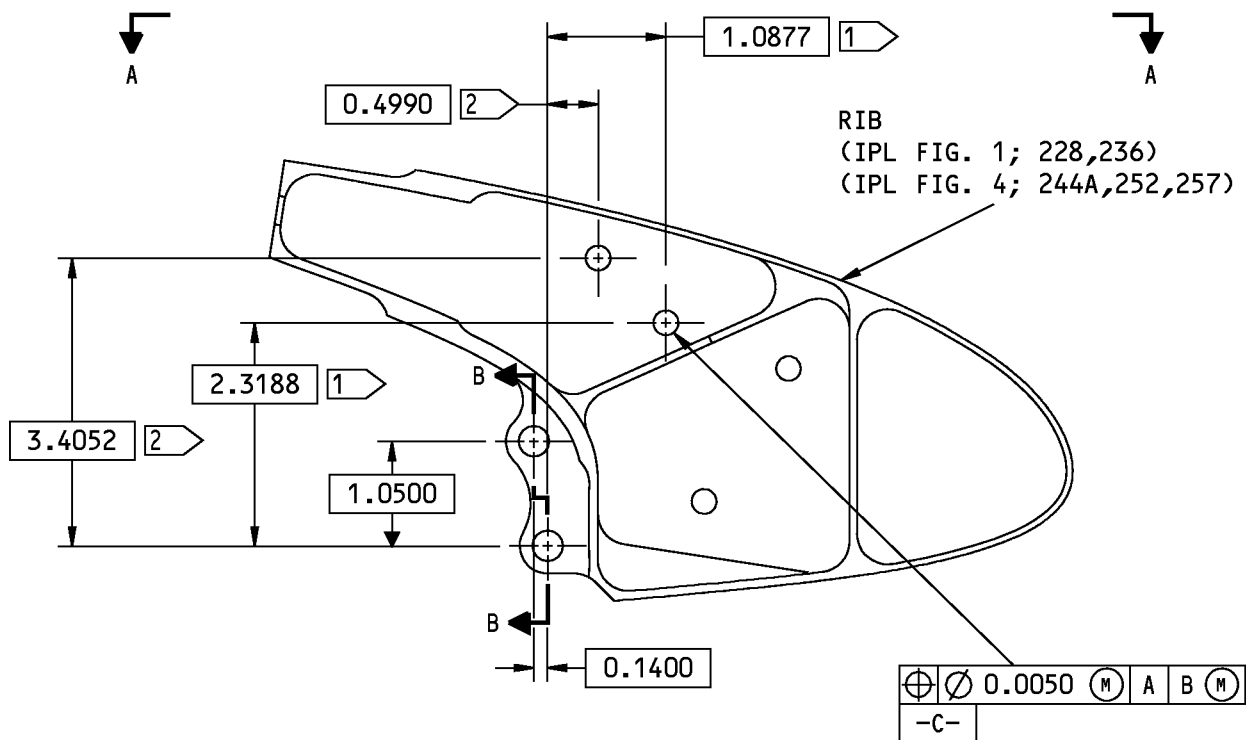
57-43-01

REPAIR 6-2

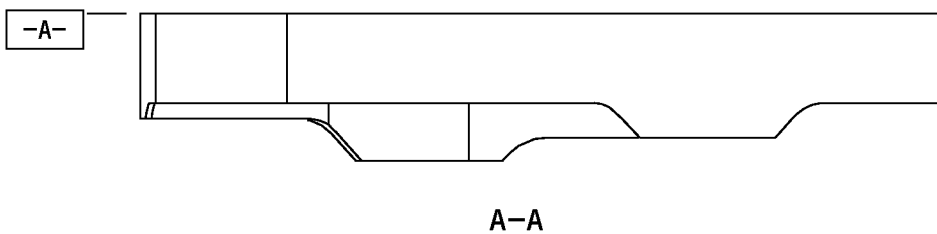
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114A7411-3,-7; 114A7412-3,-203 SHOWN
114A7411-4,-8; 114A7412-4,-204 OPPOSITE



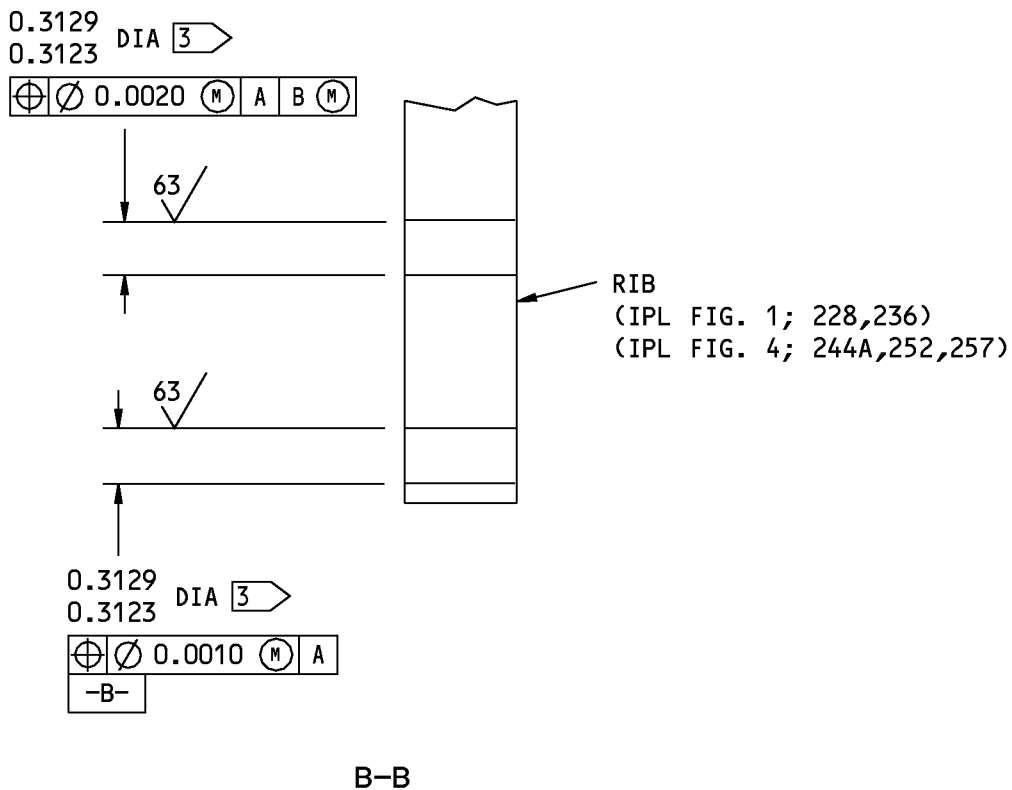
114A7411-3, -4, -7, -8; 114A7412-3, -4, -203, -204 Rib Repair
Figure 601 (Sheet 1 of 2)

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REPAIR 6-2
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- 1 FOR 114A7411-3,-4,-7,-8
- 2 FOR 114A7412-3,-4,-203,-204
- 3 DO NOT APPLY PRIMER TO THESE HOLES

125 / ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ALL DIMENSIONS ARE IN INCHES

114A7411-3, -4, -7, -8; 114A7412-3, -4, -203, -204 Rib Repair
Figure 601 (Sheet 2 of 2)

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REPAIR 6-2
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OUTBOARD/INBOARD MAIN TRACK ASSEMBLY - REPAIR 7-1

114A7511-1, -5, -7, -201, 114A7512-1, -5, -7, -201

1. General

- A. This procedure has the data necessary to repair and refinish the outboard/inboard main track assembly (IPL Figure 1; 44, 48) (IPL Figure 4; 24, 28).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subject identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.

2. Bushing Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure (REPAIR 7-1, Figure 601 and REPAIR 7-1, Figure 602

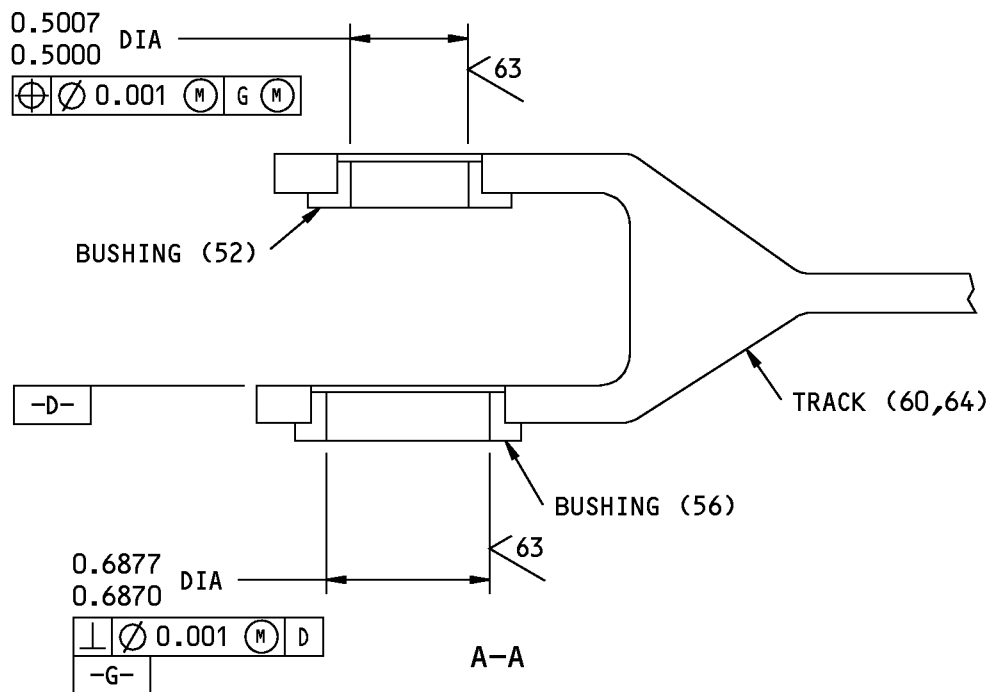
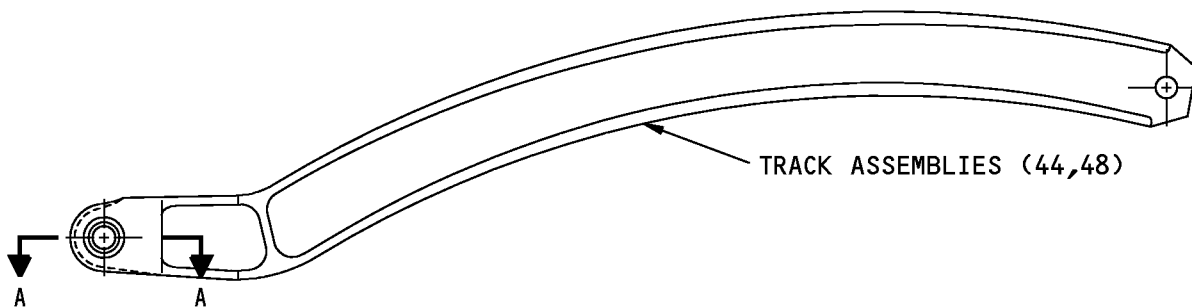
NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bushings (IPL Figure 1; 52, 56) (IPL Figure 4; 32, 36) from the track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44).
- (2) Install the bushings (IPL Figure 1; 52, 56) (IPL Figure 4; 32, 36) on the track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03) .
- (3) Ream the inside diameter of bushings (IPL Figure 1; 52, 56) (IPL Figure 4; 32, 36) to the dimensions shown on REPAIR 7-1, Figure 601 or REPAIR 7-1, Figure 602.
- (4) Break all sharp edges (0.020-0.030 inch radius). Visual check acceptable.
- (5) Fabricate oversized bushing per REPAIR 7-1, Figure 603

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REPAIR 7-1
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NOTE: SEE REPAIR 7-2, FIGURE 601 FOR BUSHING HOLE LOCATIONS

125 ✓ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

114A7511-1; 114A7512-1 Inboard and Outboard Main Track Assembly Repair
Figure 601

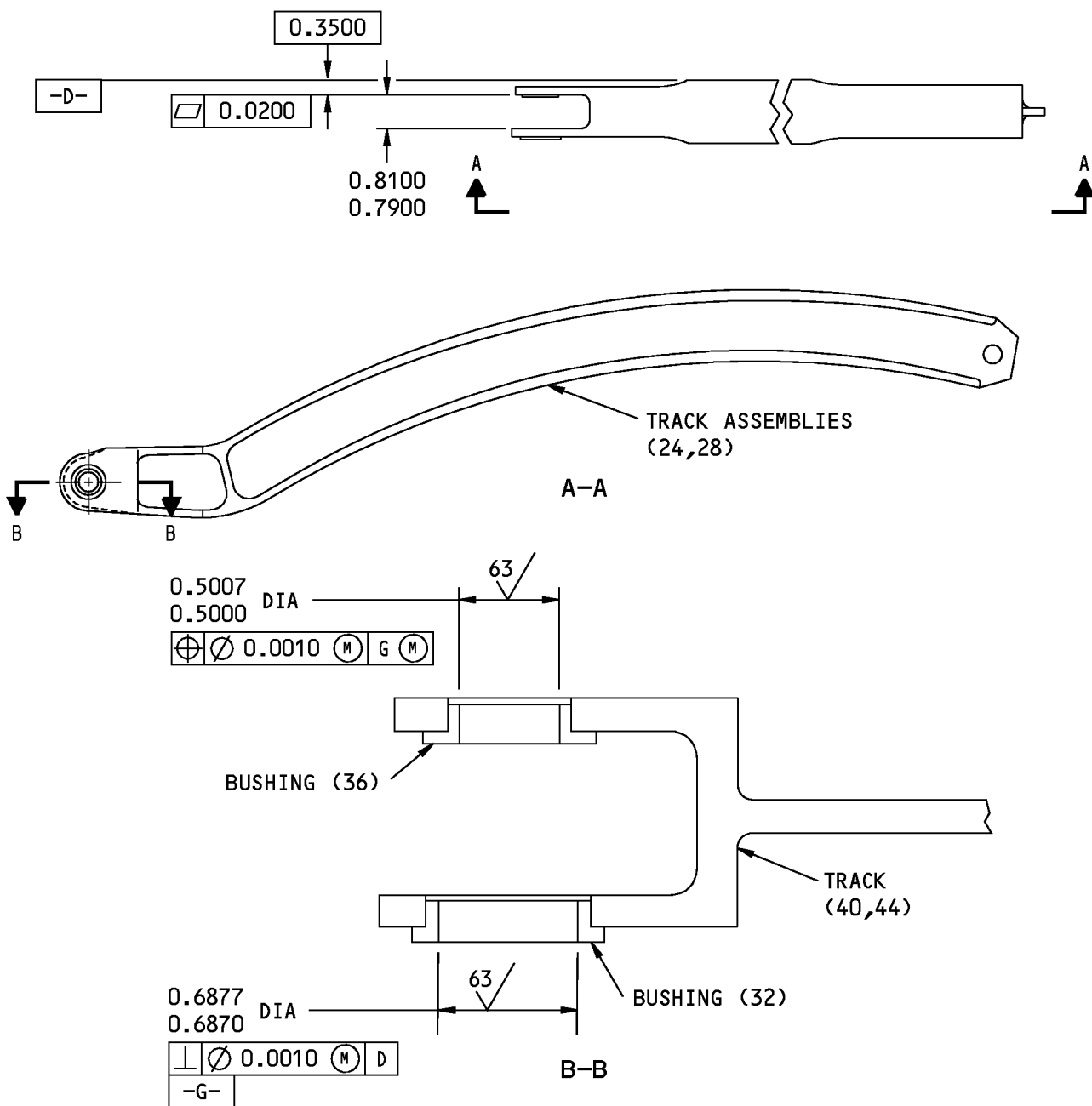
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REPAIR 7-1

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125 / ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

NOTE: SEE REPAIR 7-2, FIGURE 602 FOR BUSHING HOLE LOCATIONS

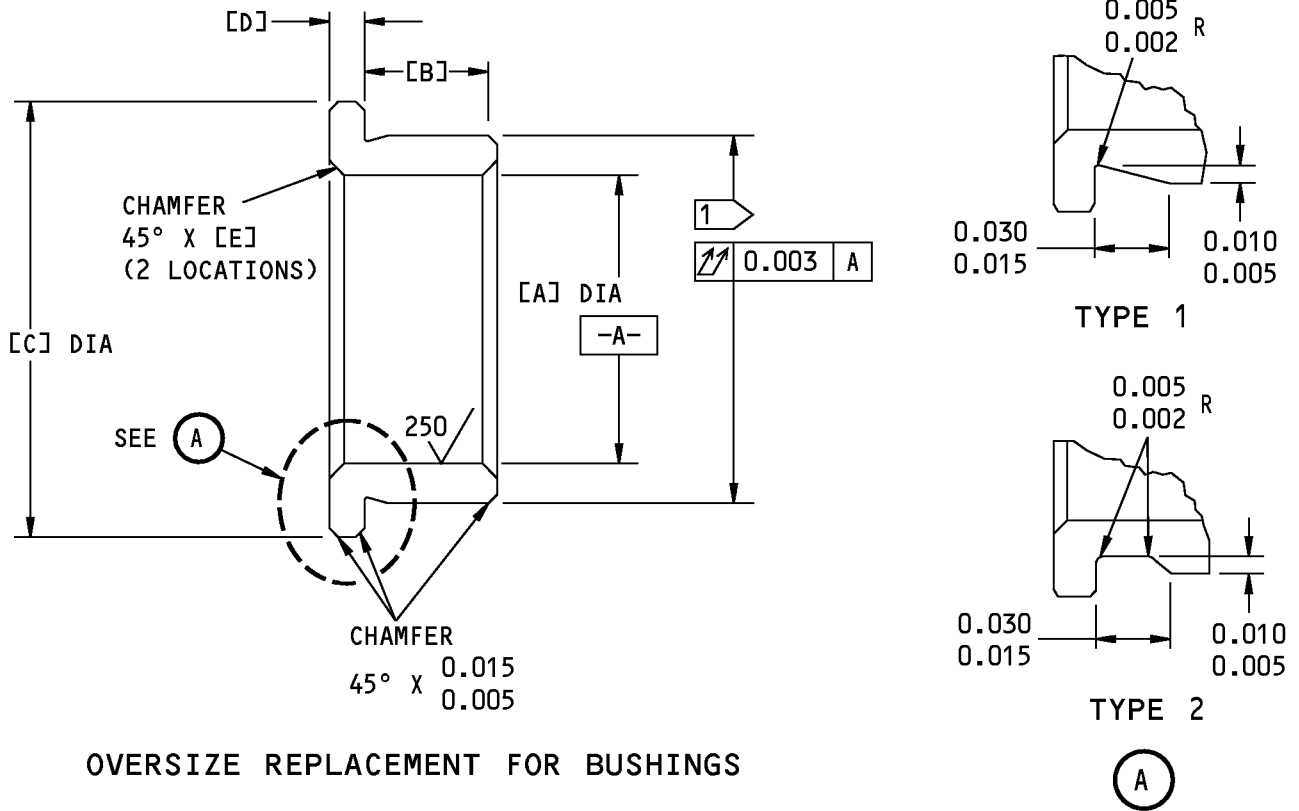
ITEM NUMBERS REFER TO IPL FIG. 4
ALL DIMENSIONS ARE IN INCHES

114A7511-5, -7, -201; 114A7512-5, -7, -201 Inboard and Outboard Main Track Assembly Repair Figure 602

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OVERSIZE REPLACEMENT FOR BUSHINGS

REPLACES BUSHING	[A]	[B]	[C]	[D]	[E]	INTER-FERENCE
IPL FIG. 1; 56 IPL FIG. 4; 32	0.678 0.672	0.180 0.175	0.960 0.950	0.074 0.069	0.025 0.015	0.0019 0.0007
IPL FIG. 1; 52 IPL FIG. 4; 36	0.491 0.484	0.180 0.175	0.870 0.860	0.065 0.060	0.025 0.015	0.0016 0.0005

1 THE OUTSIDE DIAMETER OF THE BUSHING IS EQUAL TO THE BUSHING HOLE INSIDE DIAMETER PLUS INTERFERENCE

63/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY
ALL DIMENSIONS ARE IN INCHES

Oversize Bushing Details
Figure 603

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TRACK - REPAIR 7-2

114A7511-2, -6, -8, -202, 114A7512-2, -6, -8, -202

1. General

- A. This procedure has the data necessary to repair and refinish the track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44).
- B. Refer to the Standard Overhaul Practice Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to the REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 and IPL Figure 4 for item numbers.
- E. General repair details:
 - (1) Materials: 4340M steel
Heat treat 275-300 ksi
 - (2) Shot peen: All repaired surfaces
Hard shot Rc 55-65
Intensity 0.014A-0.019A
Coverage 2.0

2. Track Refinish

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00033	Coating - Exterior Protective Enamel, Flexibility Use	BMS10-60, Type II
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

- B. References

Reference	Title
SOPM 20-10-03	SHOT PEENING
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

- C. Procedure (REPAIR 7-2, Figure 601 and REPAIR 7-2, Figure 602)

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For stripping of protective finishes, refer to SOPM 20-30-02. For finishing materials, refer to SOPM 20-60-02. For shot peening, refer to SOPM 20-10-03.

NOTE: Super detonation gun is sole source procedure. It must be done only by the authorized supplier. Also any coating still on the track must be removed by grinding or chemical stripping.

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- (1) On surfaces of track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44) indicated by flagnote 8: Apply tungsten carbide coating with a super detonation gun (F-15.360) or (F15-3871 for P/N 114A7511-8, -202 and 114A7512-8, -202) after shot peening, final coating thickness 0.004-0.006 inch. Maximum surface roughness 150 microinches Ra. Tungsten carbide overspray is allowed on surfaces indicated by flagnote 10. No tungsten carbide overspray allowed on surfaces indicated by flagnote 11.

On surfaces indicated by flagnote 9: Tungsten carbide runout 1.50 maximum. Taper must be gradual and must not terminate with a sharp edge. Minimum taper length 0.025 inch.
- (2) For P/N 114A7511-2 and 114A7512-2, do the following:
 - (a) Cad-titanium plate (F-15.32) all surfaces of the track (IPL Figure 1; 60, 64) other than the bushing holes.

NOTE: Overlap Cad-titanium plate and primer (F-19.47) 0.02 min to full coverage over the tungsten carbide coated area.
 - (b) If applicable, remove loose Cad-titanium plate in the overlap areas.
- (3) For P/N 114A7511-6, -8, -202 and 114A7512-6, -8, -202, do the following:
 - (a) Cad-Titanium plate (F-15.01) all surfaces of the track (IPL Figure 1; 60A, 64A) (IPL Figure 4; 40, 44) other than the bushing holes.

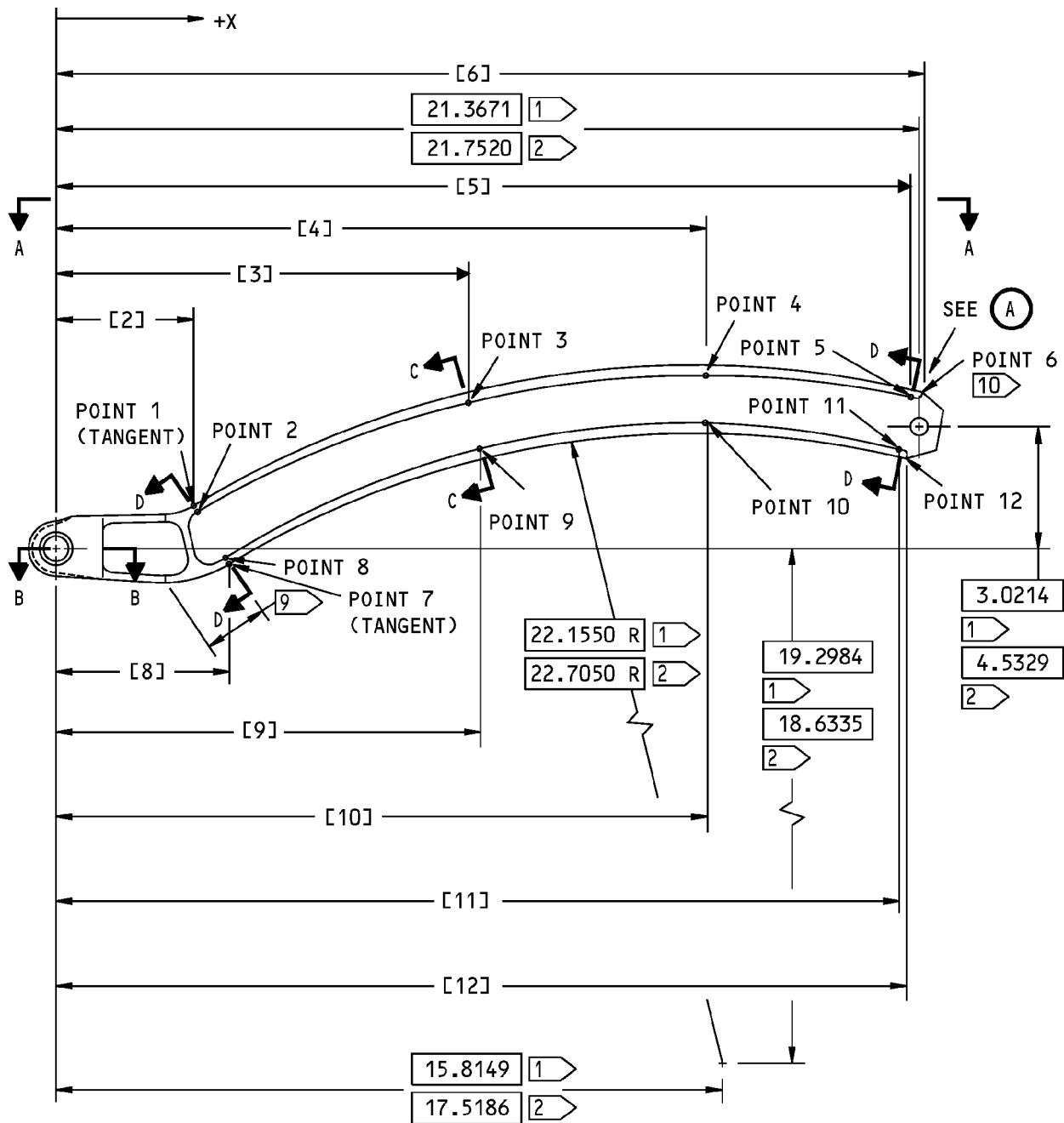
NOTE: Overlap Cad-titanium plate and primer (F-19.47) 0.02 min to full coverage over the tungsten carbide coated area.
 - (b) If applicable, remove loose Cad-titanium plate in the overlap areas.
- (4) Apply primer, C00175 (F-19.47) to all surfaces of the track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44) other than the bushing holes.

NOTE: Primer is required on the runout taper areas as shown in REPAIR 7-2, Figure 601 and 602. Overlap Cad-titanium plate and primer, C00175 (F-19.47) 0.02 min to full coverage over the tungsten carbide coated area.
- (5) Apply enamel coating, C00033 (F-19.39-707) to all surfaces of the track (IPL Figure 1; 60, 64) (IPL Figure 4; 40, 44) other than the bushing holes, 0.4370-0.4420 in hole and the tungsten carbide coated surface.

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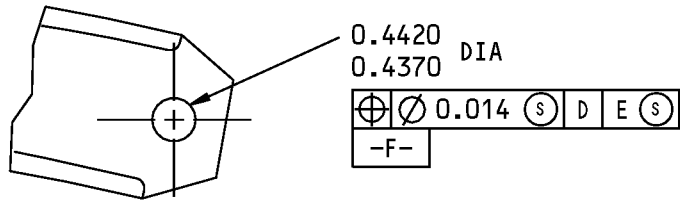
114A7511-2; 114A7512-2 Track Repair
Figure 601 (Sheet 1 of 5)

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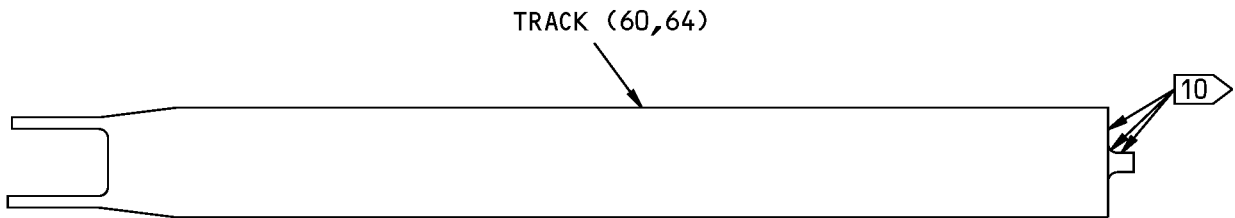
REPAIR 7-2
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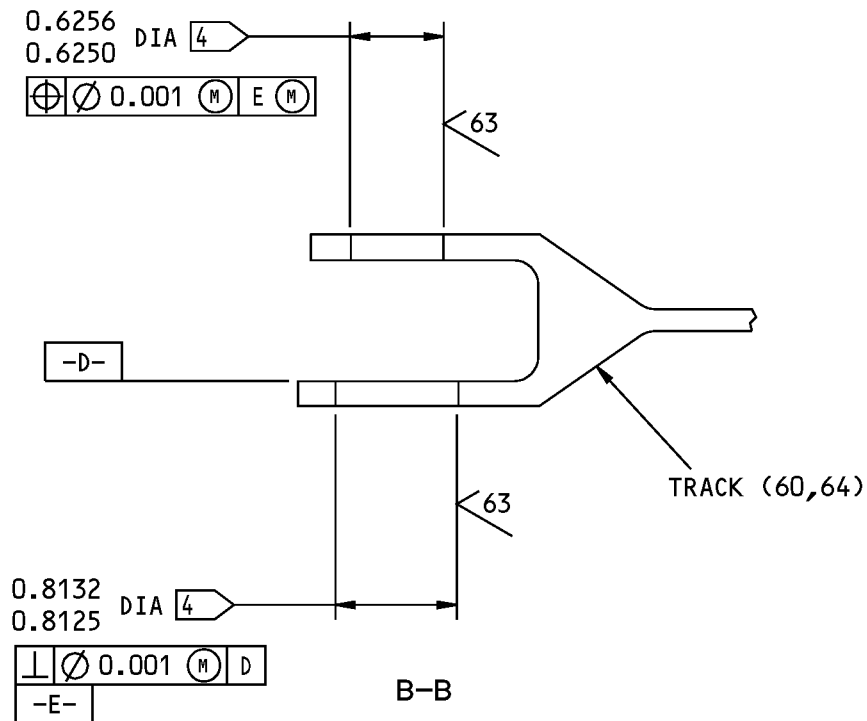
COMPONENT MAINTENANCE MANUAL



(A)



A-A



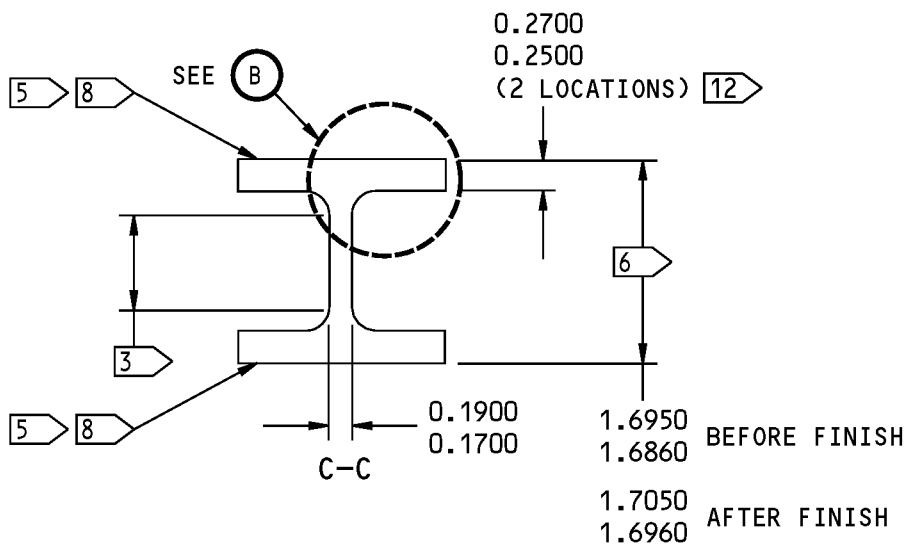
114A7511-2; 114A7512-2 Track Repair
Figure 601 (Sheet 2 of 5)

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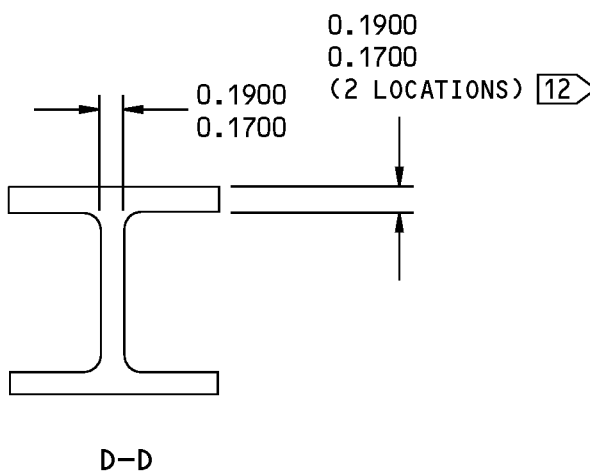
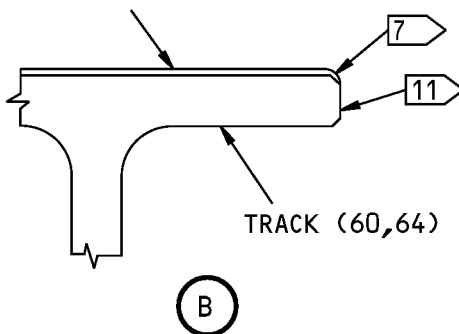
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COMPONENT MAINTENANCE MANUAL



TUNGSTEN CARBIDE COATING



114A7511-2; 114A7512-2 Track Repair
Figure 601 (Sheet 3 of 5)

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COMPONENT MAINTENANCE MANUAL

POINT	LOCATION +X	
	114A7511	114A7512
[2]	3.5307 3.4707	3.3541 3.2941
[3]	10.2386 10.1786	10.0300 9.9700
[4]	16.1190 16.0590	15.8300 15.7700
[5]	21.1961 21.1361	21.2800 21.2200
[6]	21.4582 21.4382	21.8100 21.7900
[8]	4.2228 4.1628	4.1337 4.0737
[9]	10.5167 10.4567	10.4300 10.3700
[10]	16.1054 16.0454	15.9300 15.8700
[11]	20.8953 20.8353	21.0300 20.9700
[12]	21.0600 21.0400	21.5100 21.4900

ALL DIMENSIONS ARE IN INCHES

114A7511-2; 114A7512-2 Track Repair
Figure 601 (Sheet 4 of 5)

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REPAIR 7-2

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COMPONENT MAINTENANCE MANUAL

- 1 > FOR 114A7511-2
- 2 > FOR 114A7512-2
- 3 > MACHINE FINISH 63 MICROINCHES
RA ON BOTH SIDES OF WEB.
- 4 > NO F-19.47 AND F-19.39-707 ON
BORE SURFACE.
- 5 > APPLICABLE BETWEEN POINTS 1 AND
6, 7 AND 12.
- 6 > CONSTANT HEIGHT BETWEEN POINTS
1 AND 6, 7 AND 12.
- 7 > TUNGSTEN CARBIDE RUNOUT ON BROKEN
EDGE, TAPER TO ZERO THICKNESS.
NO SHARP EDGES ALLOWED. F-19.47
REQUIRED ON RUNOUT TAPER.
F-19.39-707 OVERSPRAY PERMITTED
ON RUNOUT TAPER.
- 8 > F-15.360 AFTER SHOT PEENING.
150 MICROINCHES RA MAXIMUM.
F-15.32 AFTER COATING.
NO F-19.47 AND F-19.39-707
ALLOWED ON THIS SURFACE.
- 9 > TUNGSTEN CARBIDE RUNOUT 1.50
MAXIMUM. GRADUAL TAPER.
NO SHARP EDGES ALLOWED.
0.25 MINIMUM TAPER LENGTH FROM
POINT 1 AND POINT 7.
- 10 > TUNGSTEN CARBIDE OVERSPRAY
ALLOWED ON THIS SURFACE.
- 11 > NO TUNGSTEN CARBIDE OVERSPRAY
ALLOWED ON THIS SURFACE.
- 12 > FLANGE THICKNESS TAPERS BETWEEN
POINTS 2 AND 3, 4 AND 5, 10
AND 11, 8 AND 9.
FLANGE THICKNESS IS CONSTANT
BETWEEN POINTS 1 AND 2, 5 AND 6,
7 AND 8, 11 AND 12.

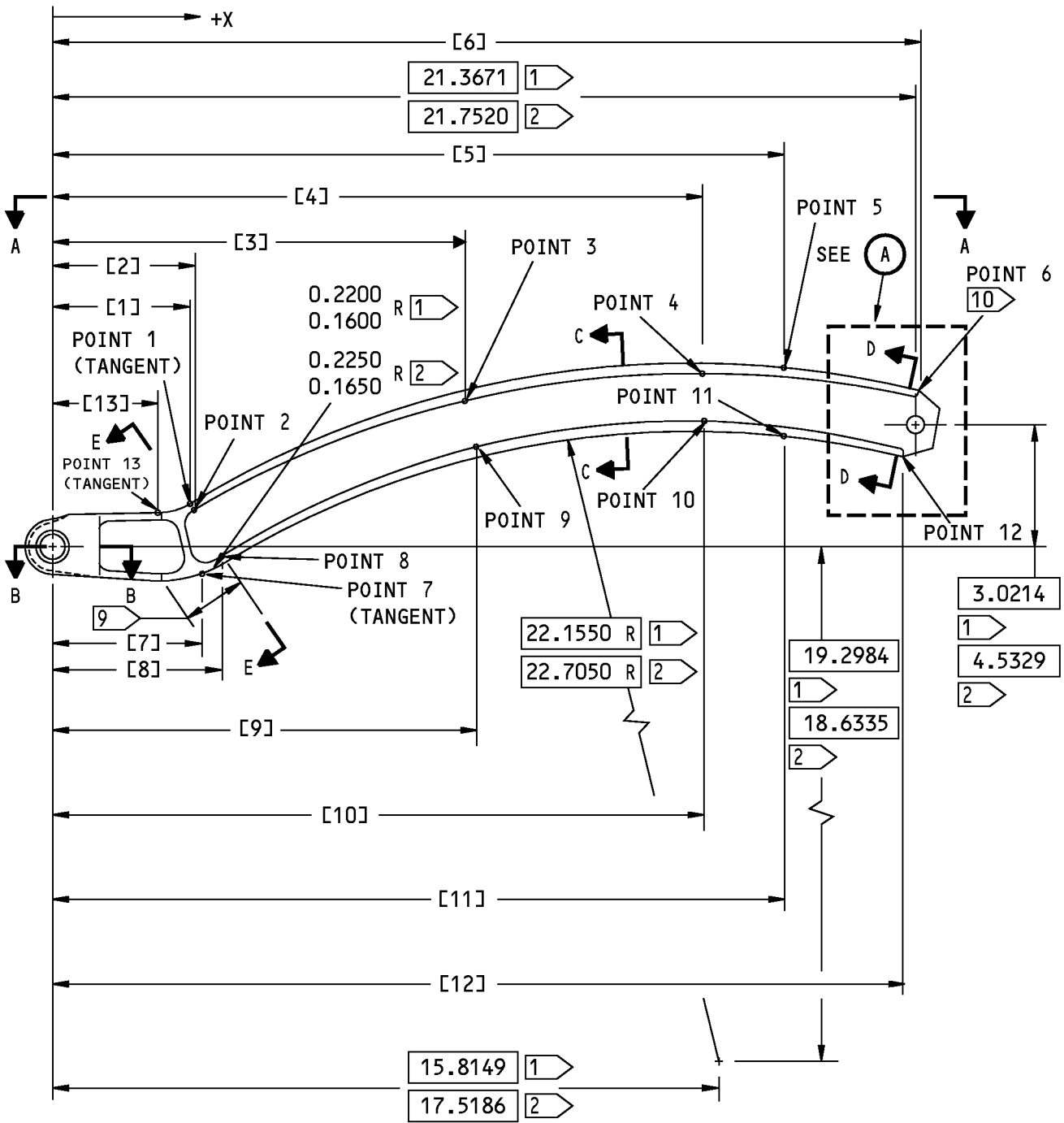
125 / ALL MACHINED SURFACES
 ITEM NUMBERS REFER TO IPL FIG. 1
 ALL DIMENSIONS ARE IN INCHES

114A7511-2; 114A7512-2 Track Repair
 Figure 601 (Sheet 5 of 5)

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COMPONENT MAINTENANCE MANUAL



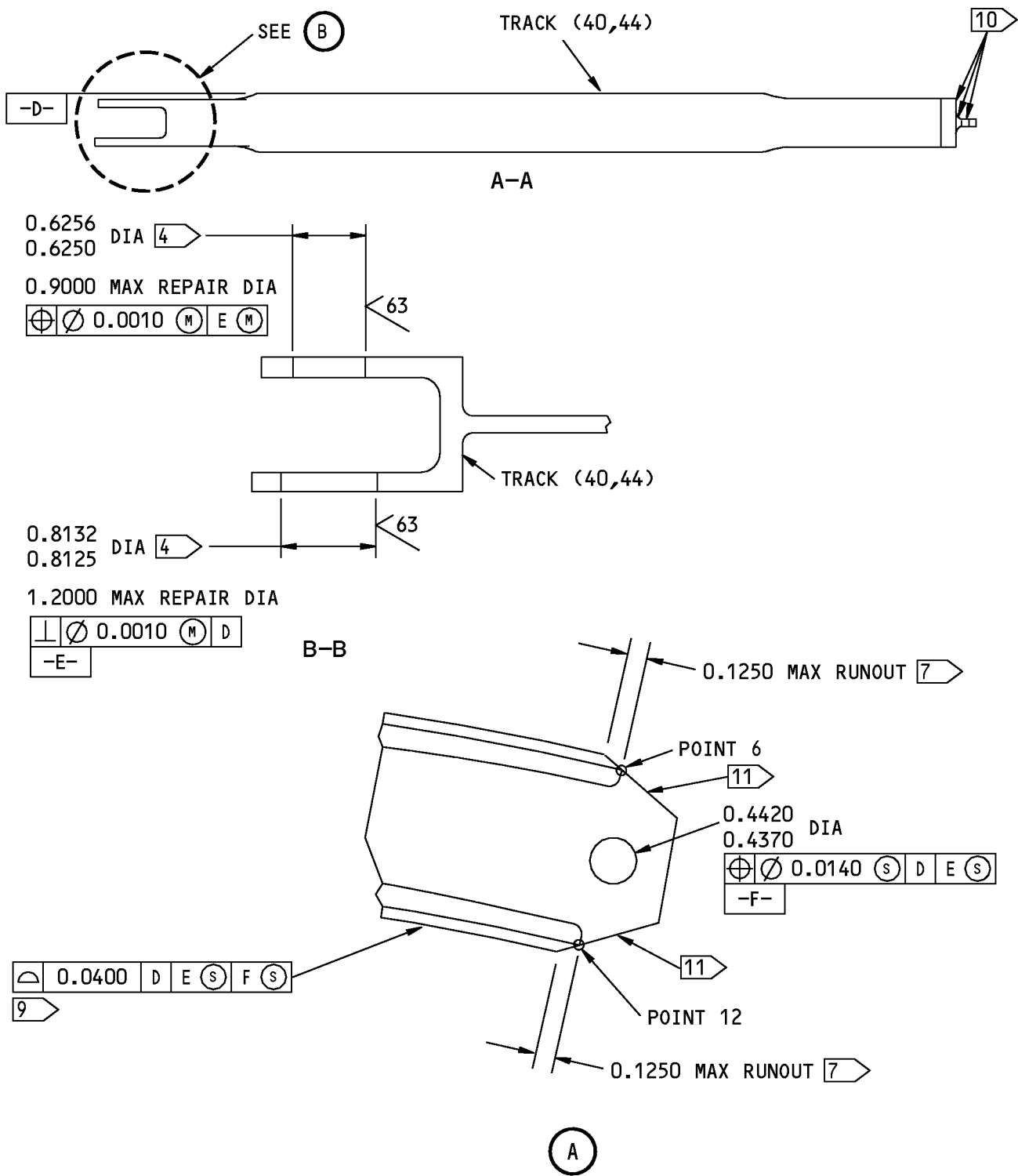
114A7511-6,-8,-202 SHOWN

114A7511-6, -8, -202; 114A7512-6, -8, -202 Track Repair
Figure 602 (Sheet 1 of 5)

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REPAIR 7-2
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COMPONENT MAINTENANCE MANUAL

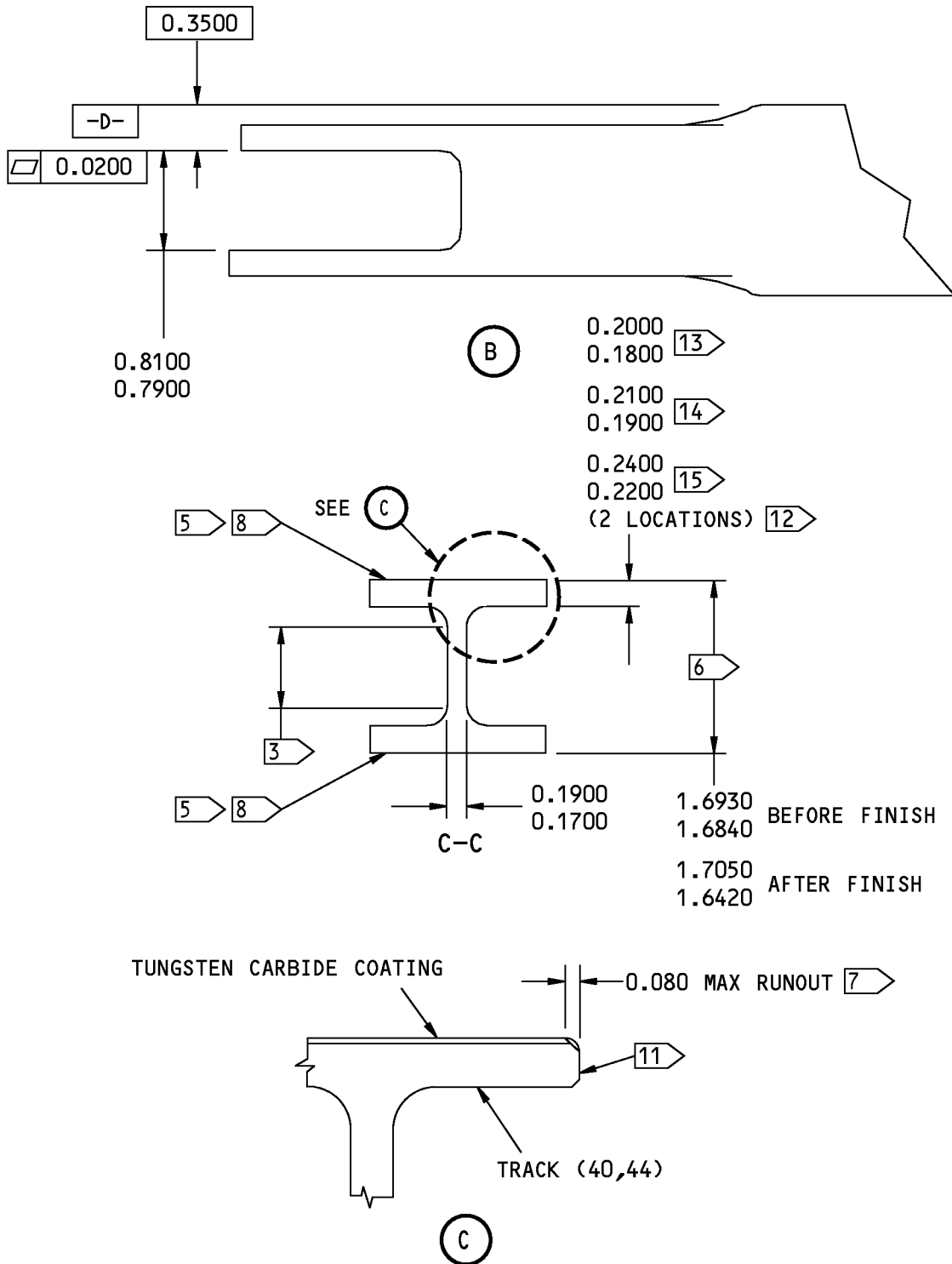


114A7511-6, -8, -202; 114A7512-6, -8, -202 Track Repair
Figure 602 (Sheet 2 of 5)

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COMPONENT MAINTENANCE MANUAL

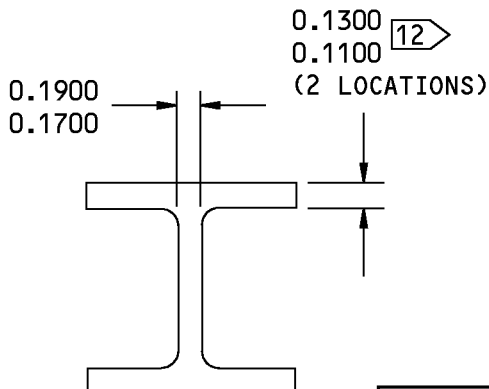


114A7511-6, -8, -202; 114A7512-6, -8, -202 Track Repair
Figure 602 (Sheet 3 of 5)

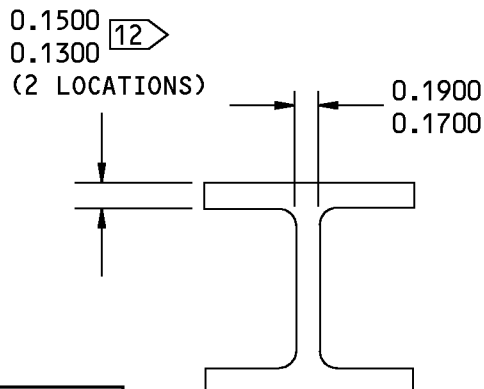
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COMPONENT MAINTENANCE MANUAL



D-D



E-E

POINT	LOCATION +X	
	114A7511	114A7512
[1]	3.5106 3.4506	3.3458 3.2858
[2]	3.5830 3.5230	3.6302 3.5702
[3]	10.9383 10.8783	10.7066 10.6466
[4]	15.4748 15.4148	14.7033 14.6433
[5]	17.9389 17.8789	17.8560 17.7960
[6]	21.4682 21.4282	21.8200 21.7800
[7]	3.6747 3.6147	3.6529 3.5929
[8]	4.3124 4.2524	4.4393 4.3793
[9]	11.2100 11.1500	11.0714 11.0114
[10]	15.4953 15.4353	14.8550 14.7950
[11]	17.8109 17.7509	17.8376 17.7776
[12]	21.0700 21.0300	21.5200 21.4800
[13]	2.7523 2.6923	2.6209 2.5609

114A7511-6, -8, -202; 114A7512-6, -8, -202 Track Repair
Figure 602 (Sheet 4 of 5)

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COMPONENT MAINTENANCE MANUAL

- 1 FOR 114A7511-6,-8,-202
- 2 FOR 114A7512-6,-8,-202
- 3 MACHINE FINISH 63 MICROINCHES RA ON BOTH SIDES OF WEB
- 4 NO F-19.47 AND F-19.39-707 ON BORE SURFACE
- 5 APPLICABLE BETWEEN POINTS 13 AND 6, 7 AND 12
- 6 CONSTANT HEIGHT BETWEEN POINTS 1 AND 6, 7 AND 12
- 7 TUNGSTEN CARBIDE RUNOUT ON BROKEN EDGE, TAPER TO ZERO THICKNESS. NO SHARP EDGES ALLOWED. F-19.47 REQUIRED ON RUNOUT TAPER. F-19.39-707 OVERSPRAY PERMITTED ON RUNOUT TAPER
- 8 F-15.360 OR F-15.3871 AFTER SHOT PEENING. 150 MICROINCHES RA MAXIMUM. F-15.01 AFTER COATING. NO F-19.39-707 ALLOWED ON THIS SURFACE
- 9 TUNGSTEN CARBIDE RUNOUT 1.50 MAXIMUM. GRADUAL TAPER. NO SHARP EDGES ALLOWED. 0.025 MINIMUM TAPER LENGTH FROM POINT 13 AND POINT 7
- 10 TUNGSTEN CARBIDE OVERSPRAY ALLOWED ON THIS SURFACE
- 11 NO TUNGSTEN CARBIDE OVERSPRAY ALLOWED ON THIS SURFACE
- 12 FLANGE THICKNESS TAPERS BETWEEN POINTS 2 AND 3, 4 AND 5, 10 AND 11, 8 AND 9. FLANGE THICKNESS IS CONSTANT BETWEEN POINTS 13 AND 2, 5 AND 6, 7 AND 8, 11 AND 12
- 13 FOR 114A7511-6,-8
- 14 FOR 114A7512-6,-8
- 15 FOR 114A7511-202; 114A7512-202

125/ ALL MACHINED SURFACES

ITEM NUMBERS REFER TO IPL FIG. 4

ALL DIMENSIONS ARE IN INCHES

114A7511-6, -8, -202; 114A7512-6, -8, -202 Track Repair
Figure 602 (Sheet 5 of 5)

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COMPONENT MAINTENANCE MANUAL

ASSEMBLY

1. General

- A. This procedure has the data necessary to assemble the wing leading edge No. 1 and 8 slat assemblies.
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to IPL Figure 1, IPL Figure 2 and IPL Figure 4 for item numbers.

2. Assembly

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I
C00528	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	MIL-C-11796, Class III

B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-01	BOLT AND NUT INSTALLATION
SOPM 20-60-02	FINISHING MATERIALS
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure (ASSEMBLY, Figure 701 and ASSEMBLY, Figure 702

NOTE: For decoding table for boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04. For finishing materials, refer to SOPM 20-60-02. For bolt and nut installation, refer to SOPM 20-50-01.

- (1) Use standard industry procedures and the steps shown below to assemble this component.
- (2) Install the bulb seals (IPL Figure 1; 760A, 764A, 776, 780, 784, 788, 792, 796, 800, 804, 808) (IPL Figure 4; 776, 780, 784, 788, 792, 796, 800, 804, 808, 812, 816, 820, 824) on the slat assembly (IPL Figure 1; 1A, 4) (IPL Figure 4; 1A, 4) with bolts (IPL Figure 1; 448, 632, 716, 720) (IPL Figure 4; 192, 460, 648, 732, 736), washers (IPL Figure 1; 452, 636, 724) (IPL Figure 4; 196, 464, 652, 740), and seal retainers (IPL Figure 1; 740A, 744, 748, 752, 756) (IPL Figure 4; 756, 760, 764, 768, 772).
- (3) Install the main track assembly (IPL Figure 1; 44, 48) (IPL Figure 4; 24, 28):
 - (a) Install the outboard main track assembly (IPL Figure 1; 48) (IPL Figure 4; 24) on the outboard main track rib assembly (IPL Figure 1; 492, 496) (IPL Figure 4; 504, 508) with bolt (IPL Figure 1; 24A) (IPL Figure 4; 8), bushing (IPL Figure 1; 68) (IPL Figure 4; 48), washers (IPL Figure 1; 32, 36) (IPL Figure 4; 12, 16A), and nut (IPL Figure 1; 40) (IPL Figure 4; 20A).

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ASSEMBLY

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COMPONENT MAINTENANCE MANUAL

- (b) Install the inboard main track assembly (IPL Figure 1; 48) (IPL Figure 4; 28) on the inboard main track rib assembly (IPL Figure 1; 500, 504) (IPL Figure 4; 512, 516) with bolt (IPL Figure 1; 24A) (IPL Figure 4; 8), bushing (IPL Figure 1; 68) (IPL Figure 4; 48), washers (IPL Figure 1; 32, 36) (IPL Figure 4; 12, 16A), and nut (IPL Figure 1; 40) (IPL Figure 4; 20A).
- (c) Install the bolt retainer (IPL Figure 1; 24A) (IPL Figure 4; 76) on the cove skin with bolts (IPL Figure 1; 8) (IPL Figure 4; 60) and washers (IPL Figure 1; 12) (IPL Figure 4; 64). Move the bolt retainer (IPL Figure 1; 24A) (IPL Figure 4; 76) along the slotted holes to make sure there is a gap between the bolt retainer (IPL Figure 1; 24A) (IPL Figure 4; 76) and the bolt head. See ASSEMBLY, Figure 701 gap dimensions.
- (4) Install the flex tabs (IPL Figure 1; 84, 88) (IPL Figure 4; 92, 96) on the slat assembly (IPL Figure 1; 1A, 4) (IPL Figure 4; 1A, 4) with bolts (IPL Figure 1; 72) (IPL Figure 4; 80), washers (IPL Figure 1; 76A) (IPL Figure 4; 84A), and nuts (IPL Figure 1; 80) (IPL Figure 4; 88).
- (5) Install the fittings (IPL Figure 1; 136, 572) (IPL Figure 4; 152, 592):
 - (a) Apply sealant, A00247 on the faying surfaces between the shim (IPL Figure 1; 140, 568) (IPL Figure 4; 156, 588) and the fitting (IPL Figure 1; 136, 572) (IPL Figure 4; 152, 592).
 - (b) Install the fitting (IPL Figure 1; 136, 572) (IPL Figure 4; 152, 592) on the rib assembly (IPL Figure 1; 200, 204, 208, 212, 656, 660, 664, 668) (IPL Figure 4; 216A, 220A, 224, 228, 229, 230, 504, 508, 512, 516, 672, 680) with shim (IPL Figure 1; 140, 568) (IPL Figure 4; 156, 588), bolts (IPL Figure 1; 124, 128, 552, 556) (IPL Figure 4; 136, 140, 564, 568), and collar (IPL Figure 1; 132, 560, 564) (IPL Figure 4; 144, 572, 580). Use as many laminations as possible between the rib lug and the fitting (IPL Figure 1; 136, 572) (IPL Figure 4; 152, 592). Maximum clamp-up gap is 0.003 inch. Use wet or dry primer, C00259 when you install the shim (IPL Figure 1; 140, 568) (IPL Figure 4; 156, 588). Remove unwanted lamination before you apply the primer, C00259.
- (6) Install the bolt retainer (IPL Figure 1; 596A, 600A) (IPL Figure 4; 616, 620) on the cove skin near the rib assembly (IPL Figure 1; 672, 676) (IPL Figure 4; 688, 692) with bolts (IPL Figure 1; 576) (IPL Figure 4; 596) and washers (IPL Figure 1; 580) (IPL Figure 4; 600).
- (7) Install the bulb seals (IPL Figure 1; 296A, 300A, 304A, 308A) (IPL Figure 4; 312A, 316A, 320A, 324A) on the cove skin with seal retainer (IPL Figure 1; 288, 292) (IPL Figure 4; 304, 308), bolts (IPL Figure 1; 260, 264) (IPL Figure 4; 276, 280), and washers (IPL Figure 1; 268) (IPL Figure 4; 284).
- (8) Put the spring retainer (IPL Figure 1; 108) (IPL Figure 4; 120), spring (IPL Figure 1; 112) (IPL Figure 4; 124), sleeve (IPL Figure 1; 104) (IPL Figure 4; 116), spacer (IPL Figure 1; 96) (IPL Figure 4; 104), on the end seal (IPL Figure 1; 116, 120) (IPL Figure 4; 128, 132) with washer (IPL Figure 1; 100, 102) (IPL Figure 4; 108, 112) and bolt (IPL Figure 1; 92) (IPL Figure 4; 100). Install the unit on the end rib assembly (IPL Figure 1; 200, 204, 208, 212) (IPL Figure 4; 216A, 220A, 224, 228) and tighten the bolts (IPL Figure 1; 92) (IPL Figure 4; 100).
- (9) Apply compound, C00528 inhibiting compound to the threads and the shank of bolts (IPL Figure 1; 368, 372) (IPL Figure 4; 384, 388). Do the next steps immediately.
- (10) Install the support (IPL Figure 1; 404, 408) (IPL Figure 4; 420, 424) and the tab (IPL Figure 1; 388, 392) (IPL Figure 4; 404, 408) on the outboard auxiliary rib assembly (IPL Figure 1; 476, 480) (IPL Figure 4; 488, 492) with bolts (IPL Figure 1; 368, 372) (IPL Figure 4; 384, 388), washers (IPL Figure 1; 376A, 380) (IPL Figure 4; 392A, 396), and nuts (IPL Figure 1; 384) (IPL Figure 4; 400). See ASSEMBLY, Figure 701 for installation dimensions.

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ASSEMBLY

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COMPONENT MAINTENANCE MANUAL

- (11) Install the support (IPL Figure 1; 412, 416) (IPL Figure 4; 428, 432) and the tab (IPL Figure 1; 396, 400) (IPL Figure 4; 412, 416) on the inboard auxiliary rib assembly (IPL Figure 1; 484, 488) (IPL Figure 4; 496, 500) with bolts IPL Figure 1; 368, 372) (IPL Figure 4; 384, 388), washers (IPL Figure 1; 376A, 380) (IPL Figure 4; 392A, 396), and nuts (IPL Figure 1; 384) (IPL Figure 4; 400). See ASSEMBLY, Figure 701 for installation dimensions.

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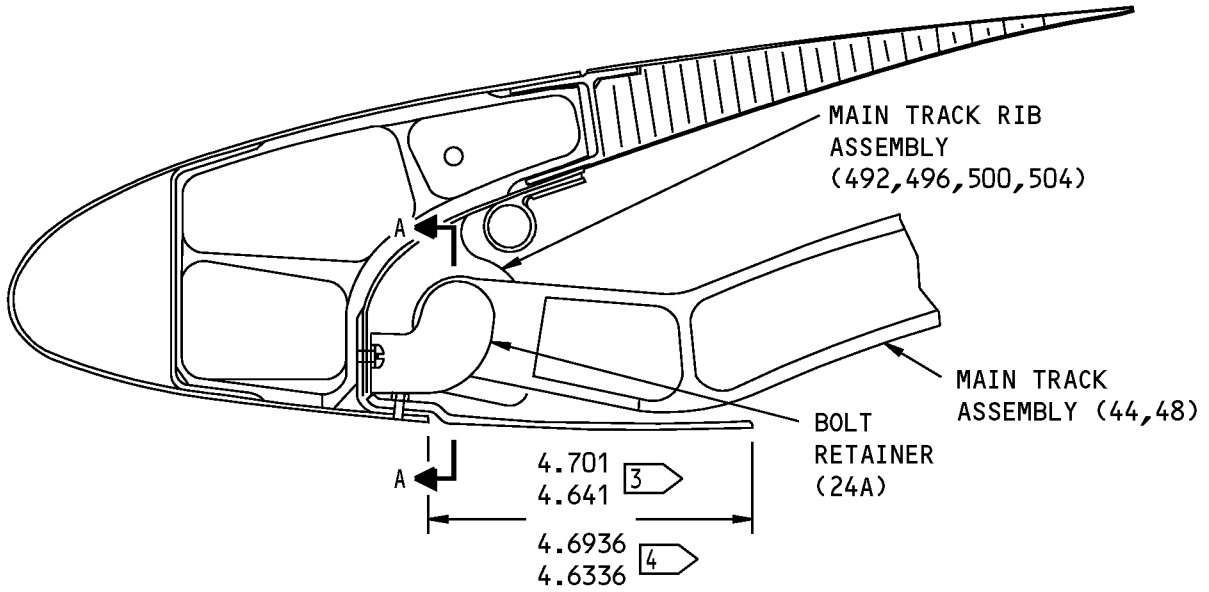
ASSEMBLY

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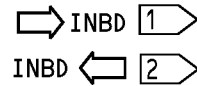
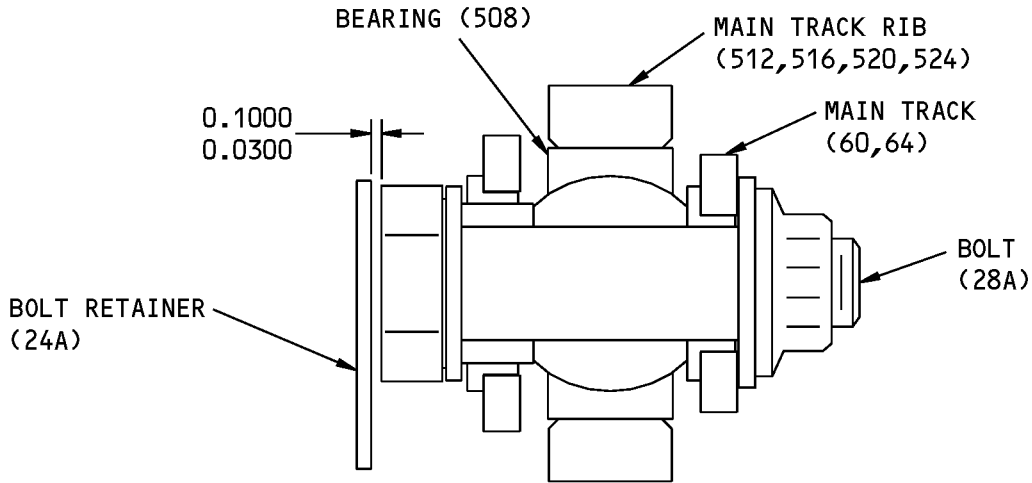
Nov 01/2006



COMPONENT MAINTENANCE MANUAL



114A7011-1 SHOWN
114A7011-2 OPPOSITE



- 1 FOR 114A5010-1,-9
- 2 FOR 114A5010-2,-10
- 3 MAIN OUTBOARD TRACK ASSEMBLY (44)
- 4 MAIN INBOARD TRACK ASSEMBLY (48)

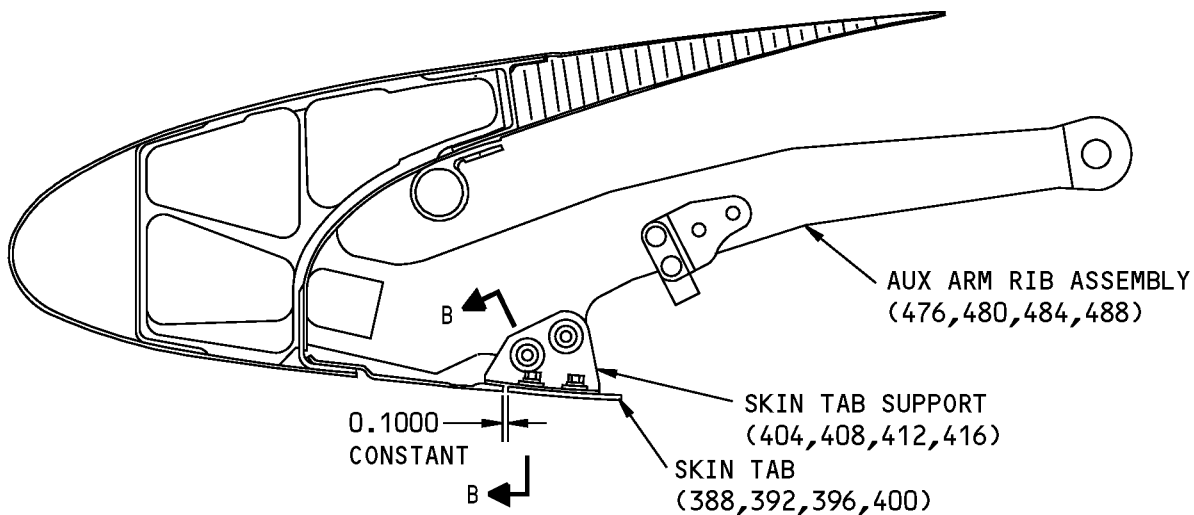
ITEM NUMBERS REFER TO IPL FIG. 1

Slat Assembly
Figure 701 (Sheet 1 of 2)

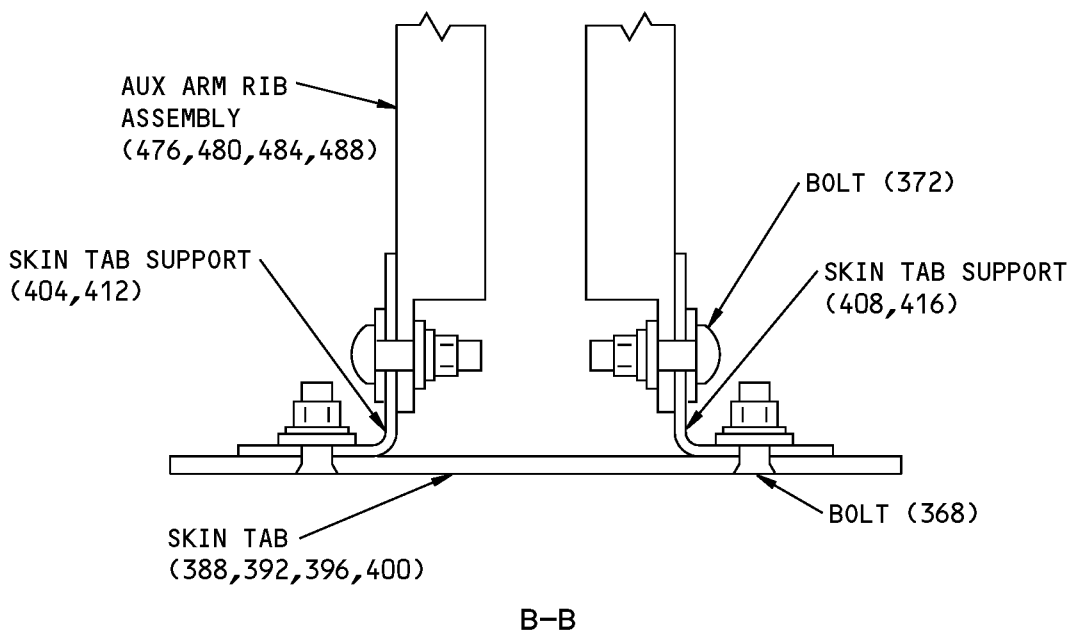
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ASSEMBLY
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COMPONENT MAINTENANCE MANUAL



114A7112-1 SHOWN
114A7112-2 OPPOSITE



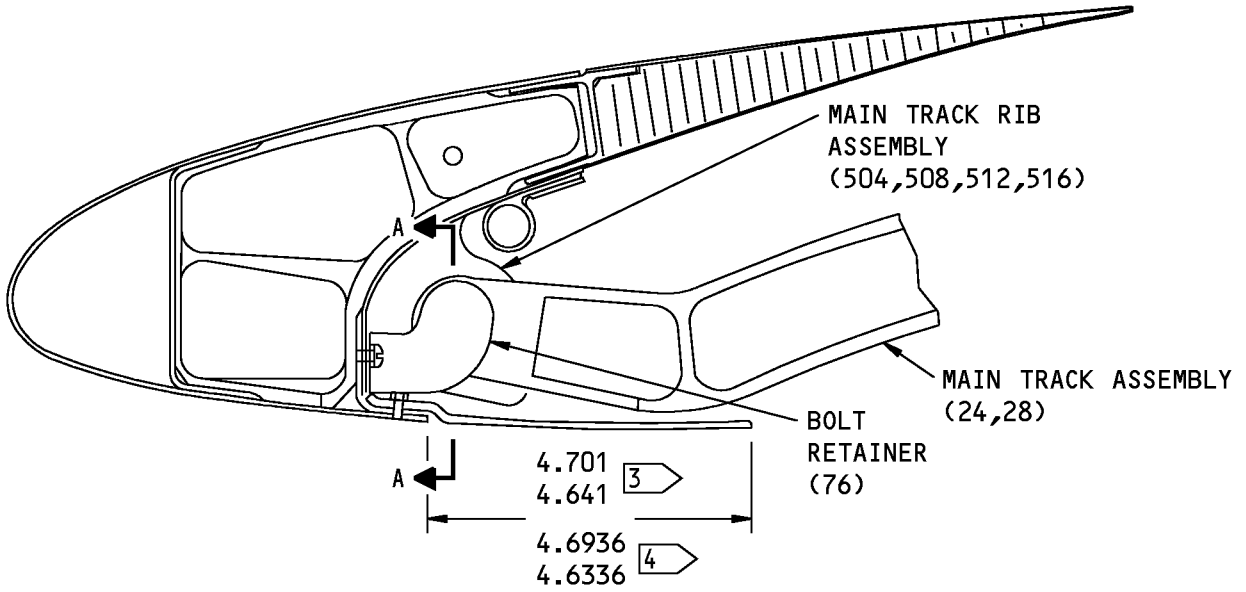
ITEM NUMBERS REFER TO IPL FIG. 1
UNLESS SHOWN DIFFERENTLY
ALL DIMENSIONS ARE IN INCHES

Slat Assembly
Figure 701 (Sheet 2 of 2)

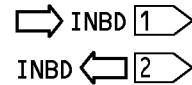
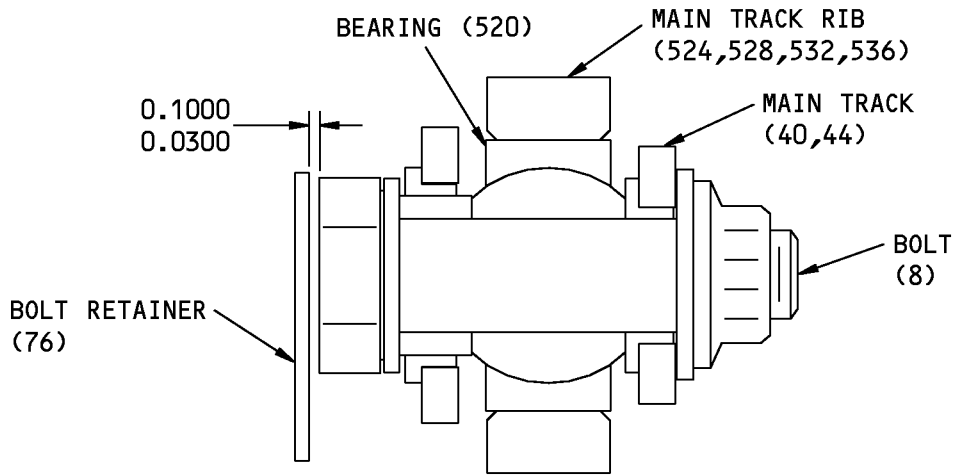
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COMPONENT MAINTENANCE MANUAL



114A7011-1 SHOWN
114A7011-2 OPPOSITE



- 1 FOR 114A5010-3,-11,-201
- 2 FOR 114A5010-4,-12,-202
- 3 MAIN OUTBOARD TRACK ASSEMBLY (24)
- 4 MAIN INBOARD TRACK ASSEMBLY (28)

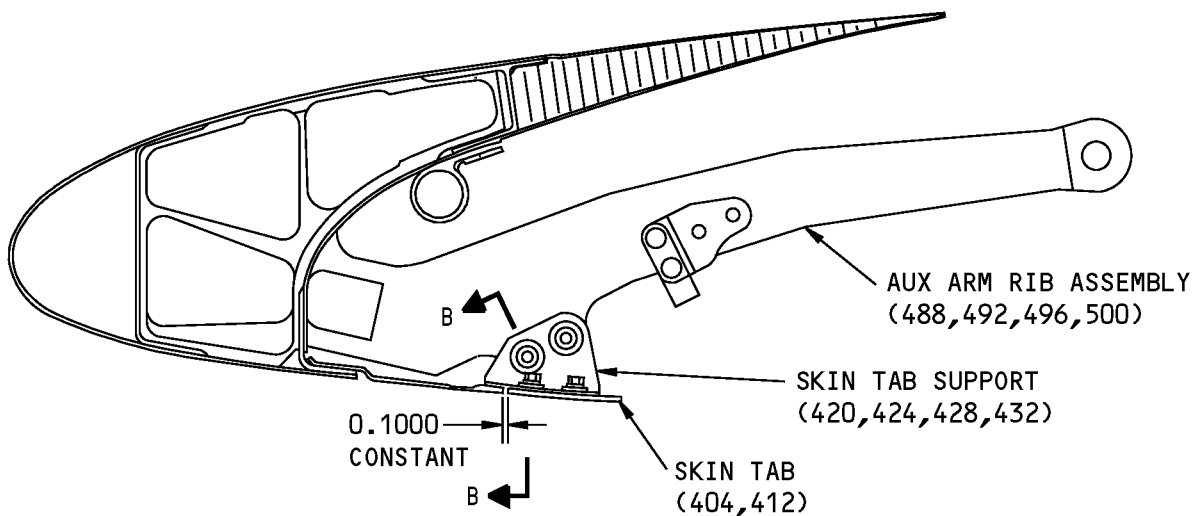
ITEM NUMBERS REFER TO IPL FIG. 4
ALL DIMENSIONS ARE IN INCHES

Slat Assembly
Figure 702 (Sheet 1 of 2)

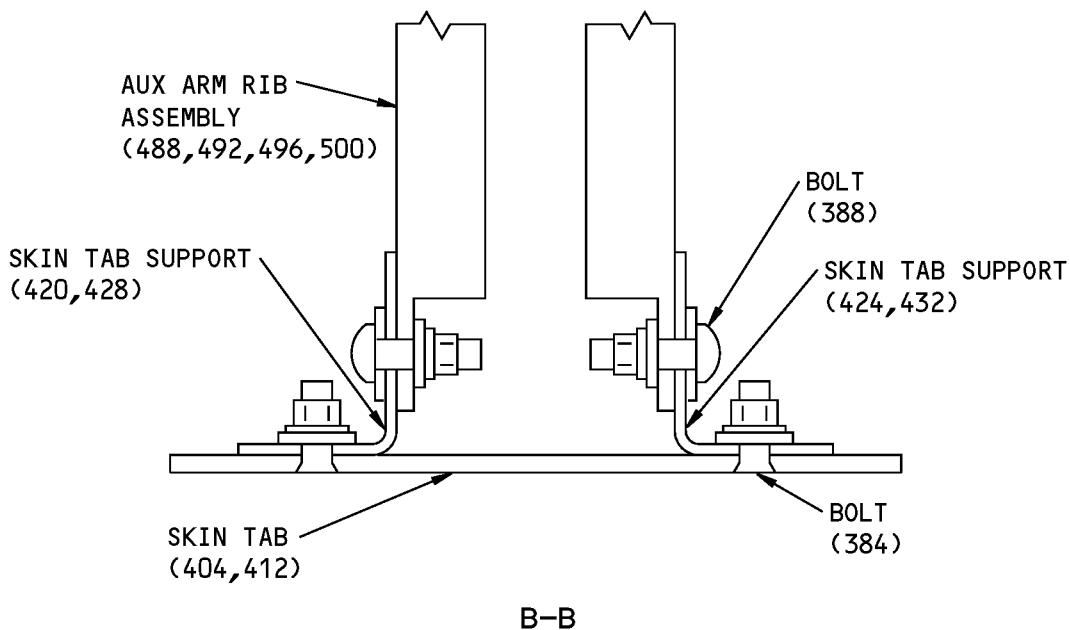
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114A7112-1 SHOWN
114A7112-2 OPPOSITE



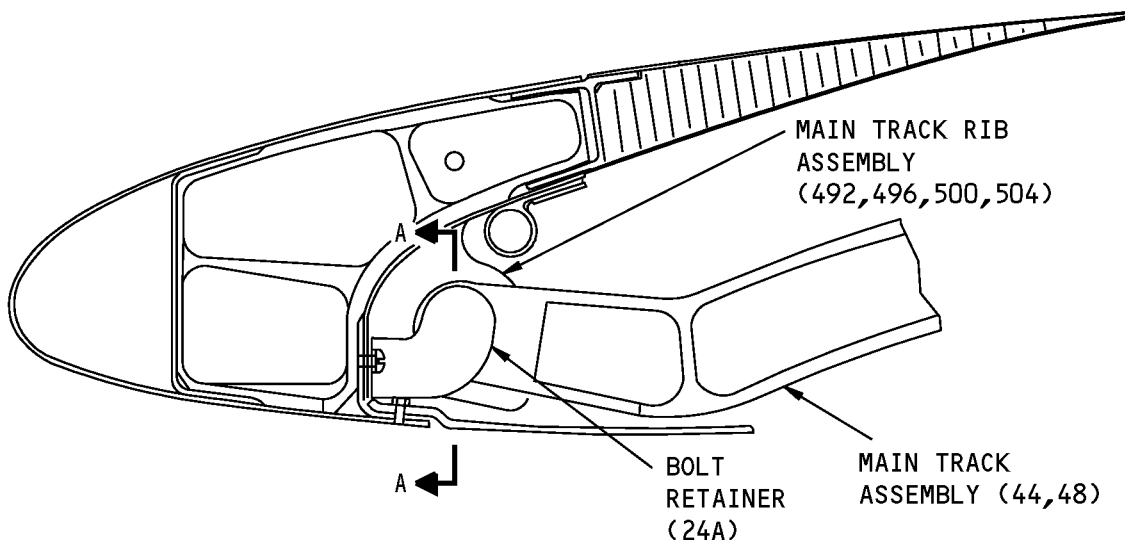
Slat Assembly
Figure 702 (Sheet 2 of 2)

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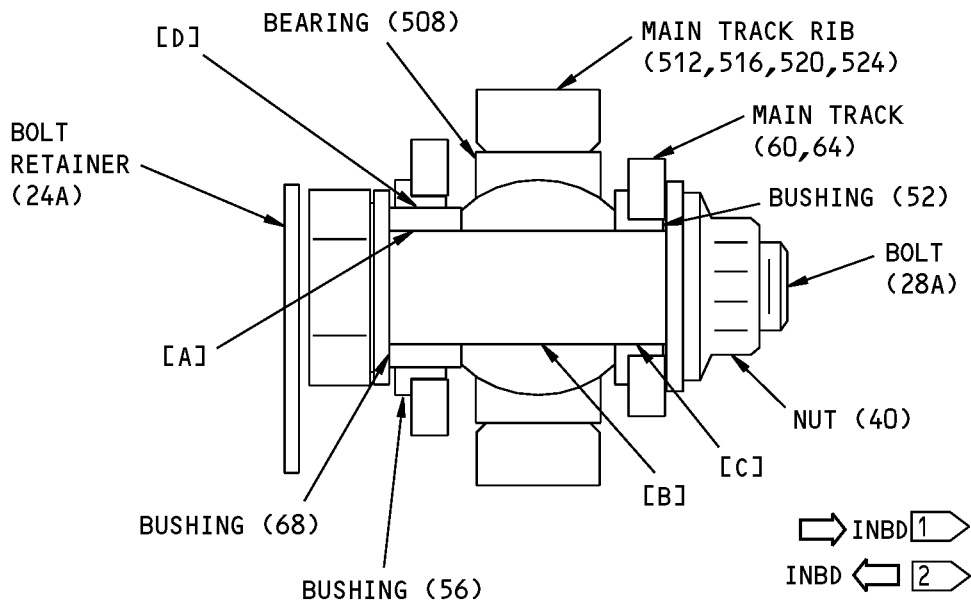
ASSEMBLY
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FITS AND CLEARANCES



114A7011-1 SHOWN 3
 114A7011-2 OPPOSITE 4



A-A

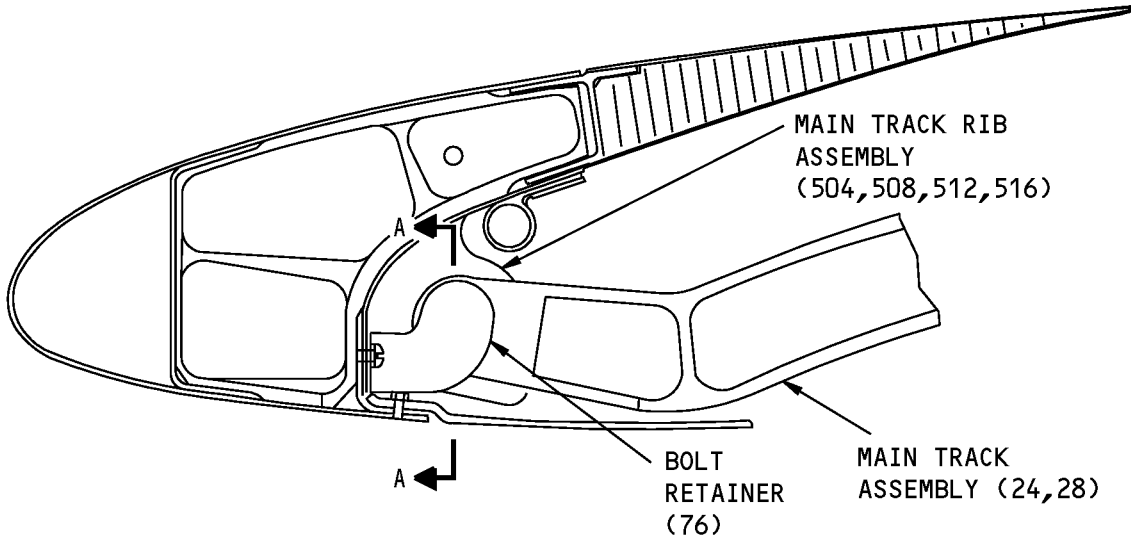
- 1 FOR 114A5010-1,-9
- 2 FOR 114A5010-2,-10
- 3 SIMILAR TO 114A7012-1
- 4 SIMILAR TO 114A7012-2

ITEM NUMBERS REFER TO IPL FIG. 1

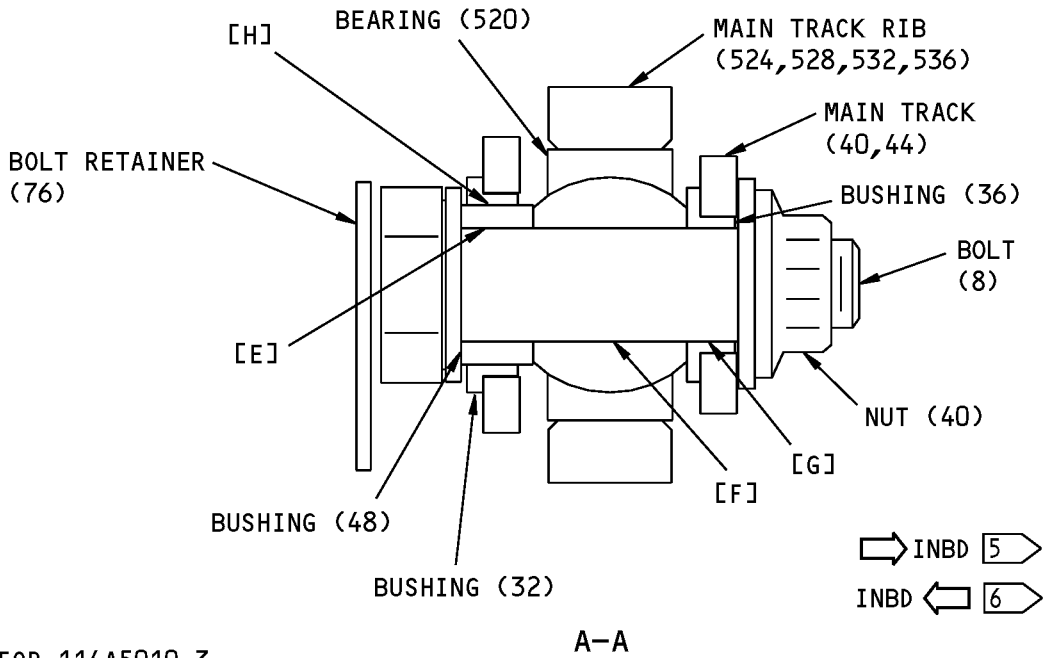
Fits and Clearances
 Figure 801 (Sheet 1 of 3)



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114A7011-1 SHOWN 7
 114A7011-2 OPPOSITE 8



- 5 FOR 114A5010-3
- 6 FOR 114A5010-4
- 7 SIMILAR TO 114A7012-1
- 8 SIMILAR TO 114A7012-2

ITEM NUMBERS REFER TO IPL FIG. 4

Fits and Clearances
 Figure 801 (Sheet 2 of 3)



COMPONENT MAINTENANCE MANUAL

REF LETTER	REF IPL		DESIGN DIMENSION*				SERVICE WEAR LIMIT*		
	FIG. NO.	MATING ITEM NO.	DIMENSION		ASSEMBLY CLEARANCE		DIMENSION		MAXIMUM CLEARANCE
			MIN	MAX	MIN	MAX	MIN	MAX	
[A]	1	ID 68	0.5000	0.5005	0.0005	0.0020	0.4970	0.5020	0.0050
		OD 28A	0.4985	0.4995					
[B]	1	ID 508	0.5000	0.5005	0.0005	0.0020	0.4970	0.5020	0.0050
		OD 28A	0.4985	0.4995					
[C]	1	ID 52	0.5000	0.5007	0.0005	0.0022	0.4970	0.5022	0.0050
		OD 28A	0.4985	0.4995					
[D]	1	ID 56	0.6870	0.6877	0.0005	0.0017	0.6845	0.6887	0.0040
		OD 68	0.6860	0.6865					
[E]	4	ID 48	0.5000	0.5005	0.0005	0.0020	0.4970	0.5020	0.0050
		OD 8	0.4985	0.4995					
[F]	4	ID 520	0.5000	0.5005	0.0005	0.0020	0.4970	0.5020	0.0050
		OD 8	0.4985	0.4995					
[G]	4	ID 36	0.5000	0.5005	0.0005	0.0022	0.4970	0.5022	0.0050
		OD 8	0.4985	0.4995					
[H]	4	ID 32	0.6870	0.6877	0.0005	0.0017	0.6845	0.6887	0.0040
		OD 48	0.6860	0.6865					

* ALL DIMENSIONS ARE IN INCHES

F55878 S00041004440_V2

Fits and Clearances
Figure 801 (Sheet 3 of 3)

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FITS AND CLEARANCES
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SPECIAL TOOLS, FIXTURES, AND EQUIPMENT

(NOT APPLICABLE)

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SPECIAL TOOLS, FIXTURES, AND EQUIPMENT

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COMPONENT MAINTENANCE MANUAL

ILLUSTRATED PARTS LIST

1. Introduction

- A. The Illustrated Parts List (IPL) contains an illustration and a list of component parts you can repair or replace. The Illustrated Parts Catalog (IPC) shows how to use the Boeing part number system.
- B. This shows how parts are related: The relation of each item to its next higher assembly (NHA) is shown in the NOMENCLATURE column. Use the indenture system that follows:

1	2	3	4	5	6	7
.	Assembly					
.	Attaching parts for assembly					
.	.	Detail parts for assembly				
.	.	Subassembly				
.	.	Attaching parts for subassembly				
.	.	.	Detail parts for subassembly			
.	.	.	Sub-subassembly			
.	.	.	Attaching parts for subassembly			
.	.	.	.	Details parts for sub-subassembly		
						Detail Installation Parts (Included only if installation parts may be sent to the shop as part of assembly)

- C. Each top assembly is given one use code letter (A, B, C, etc.) in the USAGE CODE column. All subsequent component parts in the list can have one or more of the use code letters to show effectivity to top assemblies. A component part without a use code applies to all top assemblies.
- D. An alphabetical letter is added after the item number for optional parts, parts changed by a Service Bulletin, configuration differences (except left-handed and right-handed parts), last engineering releases, and parts added between item numbers in a sequence. The alphabetical letter will not be shown on the illustration for equivalent parts of the same part number.
- E. Color-coded parts are identified with a single digit alpha following the dash number or with "SP" suffix. If the "SP" suffix is used, it represents consolidation of all color codes applicable for a given usage which are not separately listed. Orders for color-coded parts should include the registry number of the airplane for which the parts are ordered.
- F. If a part number is 15 characters long but will not fit in the part number column, the part number will be displayed with a "~" at the end of the line and will be continued on the next line. The "~" denotes that the part number continues on the next line.
- G. Parts changed by a Service Bulletin are shown by PRE SB XXXX and POST SB XXXX added to the NOMENCLATURE column.
- (1) When a new top assembly is added by a Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the top assembly level only. The configuration differences at the detail part level are shown by use code letters.
- (2) When the top assembly part number is not changed by the Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the detail level.
- H. Interchangeable Parts

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Optional (OPT)	The part is optional to and interchangeable with other parts that have the same item number.
Replaces, Replaced by and not interchangeable with (REPLACES, REPLACED BY AND NOT INTCHG/W)	The part replaces and is not interchangeable with the initial part.
Replaces, Replaced by (REPLACES, REPLACED BY)	The part replaces and is interchangeable with, or is an alternative to, the initial part.

VENDOR CODES

Code	Name
06725	AIR INDUSTRIES CORPORATION 12570 KNOTT STREET GARDEN GROVE, CALIFORNIA 92641-3932 FORMERLY AIR INDUSTRIES OF CALIF IN GARDENA, CALIF.
0PTK6	SPS TECHNOLOGIES INC AEROSPACE PRODUCTS DIV 5195 W 4700 SALT LAKE CITY, UTAH 94118 SEE V56878 SPS TECHNOLOGIES INC
15653	ALCOA GLOBAL FASTENERS INC DIV KAYNAR PRODUCTS 800 S STATE COLLEGE BLVD FULLERTON, CALIFORNIA 92831-3001 FORMERLY VK6405 MICRODOT AEROSP LTD; FORMERLY KAYNAR TECH FORMERLY FAIRCHILD FASTENERS KAYNAR DIV
15860	NEW HAMPSHIRE BALL BEARINGS, INC ASTRO DIVISION 155 LEXINGTON AVENUE LACONIA, NEW HAMPSHIRE 03246-2937 FORMERLY ASTRO BEARING CORP, LOS ANGELES, CALIF.
50632	KAMATICS CORP SUB OF KAMAN CORP 1335 BLUE HILLS ROAD BLOOMFIELD, CONNECTICUT 06002-1304
52828	REPUBLIC FASTENER MFG CORP 1300 RANCHO CONEJO BLVD NEWBURY PARK, CALIFORNIA 91320-1405 FORMERLY IN SYLMAR, CALIFORNIA

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Code	Name
53551	ALLFAST FASTENING SYSTEMS INC 15200 EAST DON JULIAN ROAD PO BOX 3166 CITY OF INDUSTRY, CALIFORNIA 91745-1001 FORMERLY V0736B FORMERLY ALLFAST INC V5K545
56878	SPS TECHNOLOGIES INC AEROSPACE AND INDUSTRIAL PRODUCTS DIV 301 HIGHLAND AVE JENKINTOWN, PENNSYLVANIA 19046 FORMERLY STANDARD PRESSED STEEL FORMERLY IN SALT LAKE, UTAH
5M902	ALCOA GLOBAL FASTENERS INC, DIV OF VOI-SHAN PRODUCTS 3000 W LOMITA BLVD TORRANCE, CALIFORNIA 90505-5103 FORMERLY FAIRCHILD INC INC FAIRCHILD AEROSPACE FASTENERS DIV
60516	WEST COAST AEROSPACE INC 812 MIRAFLORES STREET SAN PEDRO, CALIFORNIA 90731-1439
62554	SIMMONDS MECAERO FASTENERS INC 1734 SEQUOIA AVENUE ORANGE, CALIFORNIA 92668
72962	HARVARD INDUSTRIES INC 3 WERNER WAY SUITE 210 LEBANON, NEW JERSEY 08833 FORMERLY ESNA V7A079 FORMERLY ELASTIC STOP NUT IN UNION, NJ
73134	ROLLER BEARING COMPANY OF AMER DBA HEIM BEARINGS DIV 60 ROUND HILL RD FAIRFIELD, CONNECTICUT 06430-0000 FORMERLY INCOM INTL HEIM DIV; HEIM UNIVERSAL CORP INCOM; FORMERLY HEIM DIV INCOM INTL; IMO IND HEIM BEARINGS DIV
73197	HI-SHEAR TECHNOLOGY CORP 2600 SKYPARK DRIVE TORRANCE, CALIFORNIA 90509

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Code	Name
80539	SPS TECHNOLOGIES INC DIV AERPSOACE - SANTA ANA 2701 SOUTH HARBOR BOULEVARD SANTA ANA, CALIFORNIA 92704-5803 FORMERLY NUTT-SHEL DIV OF SPC WESTERN CO V80539 AND STANDARD PRESSED STEEL WESTERN DIV V17279
81376	SMITH ACQUISITION COMPANY 2240 BUENA VISTA BALDWIN PARK, CALIFORNIA 91706
85495	Replaced: [V85495] BRILES MFG CO SEE V97928 OMARK INDUSTRIES OMARK INDUSTRIES SEE PRECISION FASTENING PRECISION FASTENING SUB OF OMARK IND INC SEE DEUTSCH FASTENER CORP V08524 Replaced: [V08524] DEUTSCH FASTENER CORP SEE CODE V97928 Replaced: [V97928] HUCK INTL SEE V17446 HUCK INTL by Code: Name and Address below 17446: HUCK INTL INC AEROSPACE FASTENER DIV 900 WATSON CENTER ROAD CARSON, CALIFORNIA 90745-4201 FORMERLY V32134 REXNORD INC; FORMERLY V97928 HUCK INTL
92215	FAIRCHILD IND INC FAIRCHILD AEROSPACE FASTENER DIV 3010 W LOMITA BLVD TORRANCE, CALIFORNIA 90505-5102 FORMERLY VOI-SHAN IN CULVER CITY, CALIF
97613	SARGENT CONTROLS & AEROSPACE/KAHR BEARING DIV 5675 W BURLINGAME RD TUCSON, ARIZONA 85743 FORMERLY AETNA STEEL PROD KAHR BEARING DIV V96579 FORMERLY SARGENT IND KAHR BEARING DIV, BURBANK, CALIFORNIA
97928	Replaced: [V97928] SEE V17446 HUCK INTL by Code: Name and Address below 17446: HUCK INTL INC AEROSPACE FASTENER DIV 900 WATSON CENTER ROAD CARSON, CALIFORNIA 90745-4201 FORMERLY V32134 REXNORD INC; FORMERLY V97928 HUCK INTL
S0352	NIPPON MINIATURE BEARING CO LTD TOKYO, JAPAN

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Code	Name
U1901	BESTOBEL AVIATION PART MEGGITT COMPOSITES 127/135 FARNHAM ROAD SLOUGH BERKS SLI 4UY UNITED KINGDOM FORMERLY MEGGITT AEROSP COMPTNS BESTOBELL PROD FORMERLY VK8319

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NUMERICAL INDEX

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
10-60754-1143		1	296A	1
		4	312A	1
10-60754-1144		1	300A	1
		4	316A	1
10-60754-1145		1	304A	1
		4	320A	1
10-60754-1146		1	308A	1
		4	324A	1
102F9201M3		4	204B	2
		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
102LH9074-8		4	20A	2
109LH9074-8		1	40	2
114A5010-1		1	1A	RF
114A5010-10		1	4A	RF
114A5010-11		1	2B	RF
		4	1C	RF
114A5010-12		1	5B	RF
		4	4B	RF
114A5010-13		4	52B	1
114A5010-14		4	56B	1
114A5010-2		1	4	RF
114A5010-201		1	2A	RF
		4	1B	RF
114A5010-202		1	5A	RF
		4	4A	RF
114A5010-203		4	52A	1
114A5010-204		4	56A	1
114A5010-3		1	2	RF
		4	1A	RF
114A5010-4		1	5	RF
		4	4	RF

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
114A5010-5		4	52	1
114A5010-6		4	56	1
114A5010-9		1	1B	RF
114A5101-3		1	316	1
		4	332	1
114A5101-4		1	320	1
		4	336	1
114A5101-5		1	324	1
		4	340	1
114A5101-6		1	328	1
		4	344	1
114A5102-11		1	24A	2
		4	76	2
114A5102-17		1	596A	1
		4	616	1
114A5102-18		1	600A	1
		4	620	1
114A5102-19		4	76A	2
		4	76B	2
114A5103-1		1	364	1
		4	380	1
114A5104-1		1	396	1
		4	412	1
114A5104-2		1	400	1
		4	416	1
114A5104-3		1	388	1
		4	404	1
114A5104-4		1	392	1
		4	408	1
114A5105-10		1	412	1
		4	428	1
114A5105-11		1	408	1
		4	424	1
114A5105-12		1	404	1
		4	420	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
114A5105-9		1	416	1
		4	432	1
114A5107-1		1	140	2
		4	156	2
114A5107-2		1	568	2
		4	588	2
114A5107-4		1	360	1
		4	376	1
114A5108-1		1	348	1
		4	364	1
114A5108-2		1	344	1
		4	360	1
114A5108-3		1	352	1
		4	368	1
114A5108-4		1	356	1
		4	372	1
114A5109-1		1	843	1
		4	858	1
114A5109-2		1	844	1
		4	859	1
114A5109-7		1	843A	1
		4	858A	1
114A5109-8		1	844A	1
		4	859A	1
114A5110-201		4	900	1
114A5110-202		4	903	1
114A6010-1		1	828	1
		4	844	1
114A6010-11		4	844A	1
114A6010-12		4	848A	1
114A6010-2		1	832	1
		4	848	1
114A6010-21		4	844B	1
		4	844C	1
114A6010-22		4	848B	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	848C	1
114A7011-1		1	492	1
		4	504	1
114A7011-2		1	496	1
		4	508	1
114A7011-3		1	512	1
		4	524	1
114A7011-4		1	516	1
		4	528	1
114A7012-1		1	500	1
		4	512	1
114A7012-2		1	504	1
		4	516	1
114A7012-201		4	512A	1
114A7012-202		4	516A	1
114A7012-203		4	532A	1
114A7012-204		4	536A	1
114A7012-3		1	520	1
		4	532	1
114A7012-4		1	524	1
		4	536	1
114A7101-1		2	30	2
114A7111-1		1	476	1
		2	1A	RF
		4	488	1
114A7111-2		1	480	1
		2	5	RF
		4	492	1
114A7111-3		2	35	1
114A7111-4		2	40	1
114A7112-1		1	484	1
		2	10	RF
		4	496	1
114A7112-2		1	488	1
		2	15	RF

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	500	1
114A7112-3		2	45	1
114A7112-4		2	50	1
114A7211-1		1	672	1
		4	688	1
114A7211-2		1	676	1
		4	692	1
114A7211-201		4	688A	1
114A7211-202		4	692A	1
114A7211-203		4	704A	1
114A7211-204		4	708A	1
114A7211-3		1	688	1
		4	704	1
114A7211-4		1	692	1
		4	708	1
114A7311-1		1	656	1
		3	1A	RF
		4	672	1
114A7311-2		1	660	1
		3	5	RF
		4	676	1
114A7311-201		3	1B	RF
		4	672A	1
114A7311-202		3	5A	RF
		4	676A	1
114A7311-203		3	30A	1
114A7311-204		3	35A	1
114A7311-3		3	30	1
114A7311-4		3	35	1
114A7312-1		1	664	1
		3	10	RF
		4	680	1
114A7312-2		1	668	1
		3	15	RF
		4	684	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
114A7312-201		3	10A	RF
		4	680A	1
114A7312-202		3	15A	RF
		4	684A	1
114A7312-203		3	40A	1
114A7312-204		3	45A	1
114A7312-3		3	40	1
114A7312-4		3	45	1
114A7411-1		1	200	1
114A7411-2		1	204	1
114A7411-3		1	228	1
114A7411-4		1	232	1
114A7411-5		1	200A	1
		4	216A	1
114A7411-6		1	204A	1
		4	220A	1
114A7411-7		1	228A	1
		4	244A	1
114A7411-8		1	232A	1
		4	248A	1
114A7412-1		1	208	1
		4	224	1
114A7412-2		1	212	1
		4	228	1
114A7412-201		4	229	1
114A7412-202		4	230	1
114A7412-203		4	257	1
114A7412-204		4	258	1
114A7412-3		1	236	1
		4	252	1
114A7412-4		1	240	1
		4	256	1
114A7511-1		1	44	1
114A7511-2		1	60	1
114A7511-201		4	24A	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
114A7511-202		4	40A	1
114A7511-5		1	44A	1
		4	24	1
		4	24C	1
114A7511-6		1	60A	1
		4	40	1
114A7511-7		4	24B	1
114A7511-8		4	40B	1
114A7512-1		1	48	1
114A7512-2		1	64	1
114A7512-201		4	28A	1
114A7512-202		4	44A	1
114A7512-5		1	48A	1
		4	28	1
		4	28C	1
114A7512-6		1	64A	1
		4	44	1
114A7512-7		4	28B	1
114A7512-8		4	44B	1
114A7701-1		1	136	2
		4	152	2
114A7702-1		1	572	2
		4	592	2
114A8210-201		4	870	1
114A8210-202		4	873	1
114A8211-1		1	836	1
		4	852	1
114A8211-2		1	840	1
		4	856	1
114A8211-201		4	891	1
114A8211-202		4	894	1
114A8212-201		4	885	1
114A8212-202		4	888	1
114A9001-1		4	116A	3
114A9011-1		1	116	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	128	1
114A9011-2		1	120	1
		4	132	1
114A9101-10		1	792	1
		4	808	1
114A9101-11		1	796	1
		4	812	1
114A9101-12		1	800	1
		4	816	1
114A9101-13		1	804	1
		4	820	1
114A9101-14		1	808	1
		4	824	1
114A9101-3		1	768	1
		4	784	1
114A9101-4		1	772	1
		4	788	1
114A9101-45		1	760A	1
		4	776	1
114A9101-46		1	764A	1
		4	780	1
114A9101-5		1	776	1
		4	792	1
114A9101-6		1	780	1
		4	796	1
114A9101-7		1	784	2
		4	800	2
114A9101-9		1	788	1
		4	804	1
114A9102-2		1	744	2
		4	760	2
114A9102-22		1	288	1
		4	304	1
114A9102-23		1	292	1
		4	308	1

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
114A9102-3		1	748	2
		4	764	2
114A9102-30		1	740A	1
		4	756	1
114A9102-4		1	752	2
		4	768	2
114A9102-5		1	756	1
		4	772	1
114A9104-1		1	256	2
		4	272	2
114A9202-1		1	84	2
		4	92	2
114A9202-2		1	88	2
		4	96	2
114A9202-7		4	92A	2
114A9202-8		4	96A	2
114W4709-1		1	112	3
		4	124	3
69-77559-1		1	100	3
		4	108	3
69-77559-3		1	102	2
		4	112	2
69-77562-1		1	108	3
		4	120	3
69-77600-1		1	104	3
		4	116	3
69235-820CD		4	20A	2
69235-820CM		1	40	2
ADW08V301NC		1	508	1
		4	520	1
AF5141-3C		1	16A	8
		1	184A	22
		1	272A	16
		1	456A	88
		1	584A	4

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		1	640A	70
		1	728A	222
		4	68A	8
		4	200A	22
		4	288A	16
		4	468A	88
		4	604A	4
		4	656A	70
		4	744A	222
BA202179		1	296A	1
		4	312A	1
BA202180		1	300A	1
		4	316A	1
BA202181		1	304A	1
		4	320A	1
BA202182		1	308A	1
		4	324A	1
BACB10FA08GC		1	508	1
		4	520	1
BACB28AK08-034		1	68	2
		4	48	2
BACB28AP06P022		2	20	2
BACB28AP08P018		1	52	1
		4	36	1
BACB28AP10P032		1	680	1
		4	696	1
BACB28AT11B018C		1	56	1
		4	32	1
BACB28AT14B032C		1	684	1
		4	700	1
BACB28AW04B044C		3	20	1
BACB28AW05B044C		3	25	1
BACB28U3C030		1	224	2
		4	240	2
BACB30LH2K3		1	332A	2

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	348A	2
BACB30MR4A16		4	100A	3
BACB30MR4K16		1	92	3
		4	100	3
BACB30NR8K22		1	28A	2
		4	8	2
BACB30NT3K3		1	260	6
		1	720	59
		4	276	6
		4	736	59
BACB30NT3K4		1	372	8
		4	388	8
BACB30NT3K7		1	8	4
		1	176	11
		1	264	2
		1	448	44
		1	576	2
		1	632	35
		1	716	52
		4	60	4
		4	192	11
		4	280	2
		4	460	44
		4	596	2
		4	648	35
		4	732	52
BACB30VF3K2		4	897	2
BACB30VF3K3		1	368	8
		4	384	8
BACB30VT6K2		4	879	8
BACB30VT6K3		1	160	6
		1	436	24
		1	620	18
		4	176	6
		4	448	24

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
BACB30VU6K3		4	636	18
		1	156	4
		1	812	17
		4	172	4
		4	828	17
BACB30VU6K4		1	152	4
		1	432	26
		1	616	20
		4	168	4
		4	444	26
		4	632	20
		4	632	20
BACB30VU6K5		1	72	8
		1	148	2
		1	424	8
		1	608	6
		4	80	8
		4	164	2
		4	440	8
		4	628	6
		1	556	2
		4	568	2
BACB30YK10K16		1	556	2
		4	568	2
BACB30YK6K10		1	124	2
		4	136	2
BACB30YK6K11		1	128	2
		4	140	2
BACB30YK8K13		1	552	2
		4	564	2
BACB30YP6K5		4	80A	8
BACC30BL6		1	164	16
		1	440	58
		1	624	44
		1	816	17
		4	180	16
		4	452	56
		4	640	42
		4	640	42

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	832	17
BACC30BS6		4	882	8
BACC30CP10S		1	564	2
		4	580	2
		4	580A	2
BACC30CP6S		1	132	4
		4	144	4
		4	144A	4
BACC30CP8S		1	560	2
		4	572	2
		4	572A	2
BACC30X6S		1	132A	4
BACN10JC8CD		4	20A	2
BACN10JC8CM		1	40	2
BACN10JN3CD		4	204B	2
		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
BACN10JP3ACD		1	188	2
		1	276	6
		1	460	8
		1	644	6
		1	736	59
		4	204	2
		4	292	6
		4	472	8
		4	660	6
		4	752	59
BACN10JP4ACD		1	220	3
		4	236	3
BACN10YF31CD		4	204A	2
		4	242	2
		4	292A	6
		4	472A	8

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY	
BACN10YF32CD		4	660A	6	
		4	752A	59	
		1	20	4	
		1	472	2	
		1	588A	1	
		1	648	12	
		4	484	2	
		4	608	1	
		4	664	12	
	BACN10YF33CD		1	192	2
		1	284	1	
		1	464	27	
		1	652	17	
		1	732	52	
		4	208	2	
		4	300	1	
		4	476	27	
		4	668	17	
		4	748	52	
BACN10YF34CD			1	196	7
			1	280	1
			1	468	7
		1	592A	1	
		4	72A	4	
		4	212	7	
		4	296	1	
		4	480	7	
		4	612	1	
	BACN10YR08CD		1	340	2
		4	356	2	
BACN10YR3CD		1	80	8	
		1	384	16	
		4	88	8	
BACN10YT3CS		4	400	16	
		1	133	4	

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	148	4
BACN10YT3CSA		4	148A	4
BACN10YT4CS		1	561	2
		4	576	2
BACN10YT4CSA		4	576A	2
BACN10YT5CS		1	565	2
		4	584	2
BACN10YT5CSA		4	584A	2
BACR15BA3AD		1	216	6
		4	232	6
BACR15BA3ADC		4	241	4
BACR15BB3AD		1	845	4
		4	860	4
BACR15BB5AD		2	25	4
BACR15BB8AD		1	824	5
		4	840	5
BACR15BB8AD5C		4	876	5
BACR15DR3AC		1	16A	8
		1	184A	22
		1	272A	16
		1	456A	88
		1	584A	4
		1	640A	70
		1	728A	222
		4	68A	8
		4	200A	22
		4	288A	16
		4	468A	88
		4	604A	4
		4	656A	70
		4	744A	222
BACR15FT5AD		1	312	4
		4	328	4
BACR15FT8AD		1	144A	2
		1	420A	4

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
BACR15FT8D		1	604A	3
		4	160A	2
		4	436A	4
		4	624A	3
		1	144	2
		1	420	4
		1	604	3
		4	160	2
		4	436	4
		4	624	3
BACR15GE3CW		1	16	8
		1	184	22
		1	272	16
		1	456	88
		1	584	4
		1	640	70
		1	728	222
		4	68	8
		4	200	22
		4	288	16
		4	468	88
		4	604	4
		4	656	70
		4	744	222
BACR15GF6D		1	172	7
		1	444	30
		1	628	20
		1	820	263
		4	188	7
		4	456	30
		4	644	20
		4	836	263
BACS40R008B009F		1	248	AR
		4	264	AR
BACS40R008B018F		1	536	AR

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		1	708	AR
		4	548	AR
		4	724	AR
BACS40R008B020F		1	540	AR
		1	704	AR
		4	552	AR
		4	720	AR
BACS40R009B017F		1	532	AR
		1	700	AR
		4	544	AR
		4	716	AR
BACS40R009C010F		1	244	AR
		4	260	AR
BACS40R010C011F		1	252	AR
		4	268	AR
BACS40R010C017F		1	528	AR
		1	696	AR
		4	540	AR
		4	712	AR
BACS40R010C020F		1	544	AR
		1	712	AR
		4	556	AR
		4	728	AR
BACS40R010C030F		1	548	AR
		4	560	AR
BACW10BP8ACU		1	32	2
		4	12	2
BACW10BP8APU		1	36	2
BACW10BP8DP		4	16A	2
BACW10P42S		1	380	8
		4	396	8
BMN4122C1D2-8		1	40	2
		1	40	2
BMN4122CPD8-8		4	20A	2
BRFM20C3D		4	204B	2

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
H01-8BAC		1	40	2
H51650-8BAC		4	20A	2
H52732-08CD		1	340	2
		4	356	2
H52732-3CD		1	80	8
		1	384	16
		4	88	8
		4	400	16
HL1187DU6		1	132A	4
		1	132A	4
		1	132A	4
		1	132A	4
HL87DU6		1	132A	4
		1	132A	4
		1	132A	4
HLT422-10-16		1	556A	2
HLT422AP10-16		1	556	2
		1	556	2
		1	556	2
		4	568	2
		4	568	2
		4	568	2
HLT422AP6-10		1	124	2
		1	124	2
		1	124	2
		4	136	2
		4	136	2
		4	136	2
HLT422AP6-11		1	128	2
		1	128	2
		1	128	2

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
HLT422AP8-13		4	140	2
		4	140	2
		4	140	2
		1	552	2
		1	552	2
		1	552	2
		4	564	2
		4	564	2
		4	564	2
	HLT422PB6-11		1	128A
HLT422SE6-10		1	124A	2
HST1094KP6		4	882	8
HST10AG6-2		4	879	8
		4	879	8
		4	879	8
		4	879	8
HST10AG6-3		1	160	6
		1	160	6
		1	160	6
		1	160	6
		1	436	24
		1	436	24
		1	436	24
		1	436	24
		1	620	18
		1	620	18
		1	620	18
		1	620	18
		4	176	6
		4	176	6
		4	176	6
		4	176	6
	4	448	24	
	4	448	24	
	4	448	24	

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY	
HST11AG6-3		4	448	24	
		4	636	18	
		4	636	18	
		4	636	18	
		4	636	18	
		1	156	4	
		1	156	4	
		1	156	4	
		1	156	4	
		1	812	17	
		1	812	17	
		1	812	17	
		1	812	17	
		4	172	4	
		4	172	4	
		4	172	4	
		4	172	4	
		4	828	17	
	HST11AG6-4		4	828	17
			4	828	17
		4	828	17	
		4	828	17	
		1	152	4	
		1	152	4	
		1	152	4	
		1	152	4	
		1	432	26	
		1	432	26	
		1	432	26	
		1	432	26	
		1	616	20	
		1	616	20	
		1	616	20	
	4	168	4		
	4	168	4		

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
HST11AG6-5		4	168	4
		4	168	4
		4	444	26
		4	444	26
		4	444	26
		4	444	26
		4	632	20
		4	632	20
		4	632	20
		4	632	20
		1	72	8
		1	72	8
		1	72	8
		1	72	8
		1	148	2
		1	148	2
		1	148	2
		1	148	2
		1	424	8
		1	424	8
		1	424	8
		1	424	8
		1	608	6
		1	608	6
		1	608	6
		1	608	6
		4	80	8
		4	80	8
		4	80	8
		4	80	8
		4	164	2
		4	164	2
	4	164	2	
	4	164	2	
	4	440	8	

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
HST79-6		4	440	8
		4	440	8
		4	440	8
		4	628	6
		4	628	6
		4	628	6
		4	628	6
		1	164	16
		1	164	16
		1	164	16
		1	440	58
		1	440	58
		1	440	58
		1	624	44
		1	624	44
		1	624	44
		1	816	17
		1	816	17
		1	816	17
	HST79CY6		4	180
		4	452	56
		4	640	42
		4	832	17
		1	164	16
		1	440	58
		1	624	44
		1	816	17
		4	180	16
		4	180	16
		4	180	16
		4	452	56
		4	452	56
		4	452	56
		4	640	42
		4	640	42

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
		4	640	42
		4	832	17
		4	832	17
		4	832	17
KSC152200BZ08GC		1	508	1
		4	520	1
KWDB08-35		1	508	1
		4	520	1
MF51594-3-2BAC		1	20	4
		1	472	2
		1	588A	1
		1	648	12
		4	484	2
		4	608	1
		4	664	12
MF51594-3-3BAC		1	192	2
		1	284	1
		1	464	27
		1	652	17
		1	732	52
		4	208	2
		4	300	1
		4	476	27
		4	668	17
		4	748	52
MF51594-3-4BAC		1	196	7
		1	280	1
		1	468	7
		1	592A	1
		4	72A	4
		4	212	7
		4	296	1
		4	480	7
		4	612	1
MF51637-3		4	204B	2

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
MF53050-3CD		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
		4	204B	2
		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
	MS27253-1		1	850
		4	864	1
MS27253F2		4	864A	1
NAS1149D0316J		1	12	4
		1	180	11
		1	268	8
		1	452	44
		1	580	2
		1	636	35
		1	724	111
		4	64	4
		4	196	11
		4	284	8
		4	464	44
		4	600	2
		4	652	35
		4	740	111
NAS1149D0363J		1	76A	8
		1	376A	16
		4	84A	8
		4	392A	16
NAS1149DN816J		1	336	2
		4	352	2
NAS1399D4A2		1	842	4
		4	857	4
NAS1399D5A2		1	168	12

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
NAS42DD8A54FC		4	184	12
		1	96	3
		4	104	3
NS202487-02		4	204B	2
		4	292B	6
		4	472B	8
		4	660B	6
		4	752B	59
PLH508CD		1	340	2
		4	356	2
PLH53CD		1	80	8
		1	384	16
		4	88	8
		4	400	16
SWKRS08-350SC		1	508	1
		4	520	1
WC331K6-5		4	80A	8
WES08FAGC		1	508	1
		4	520	1
WHTFA08VC		1	508	1
		4	520	1

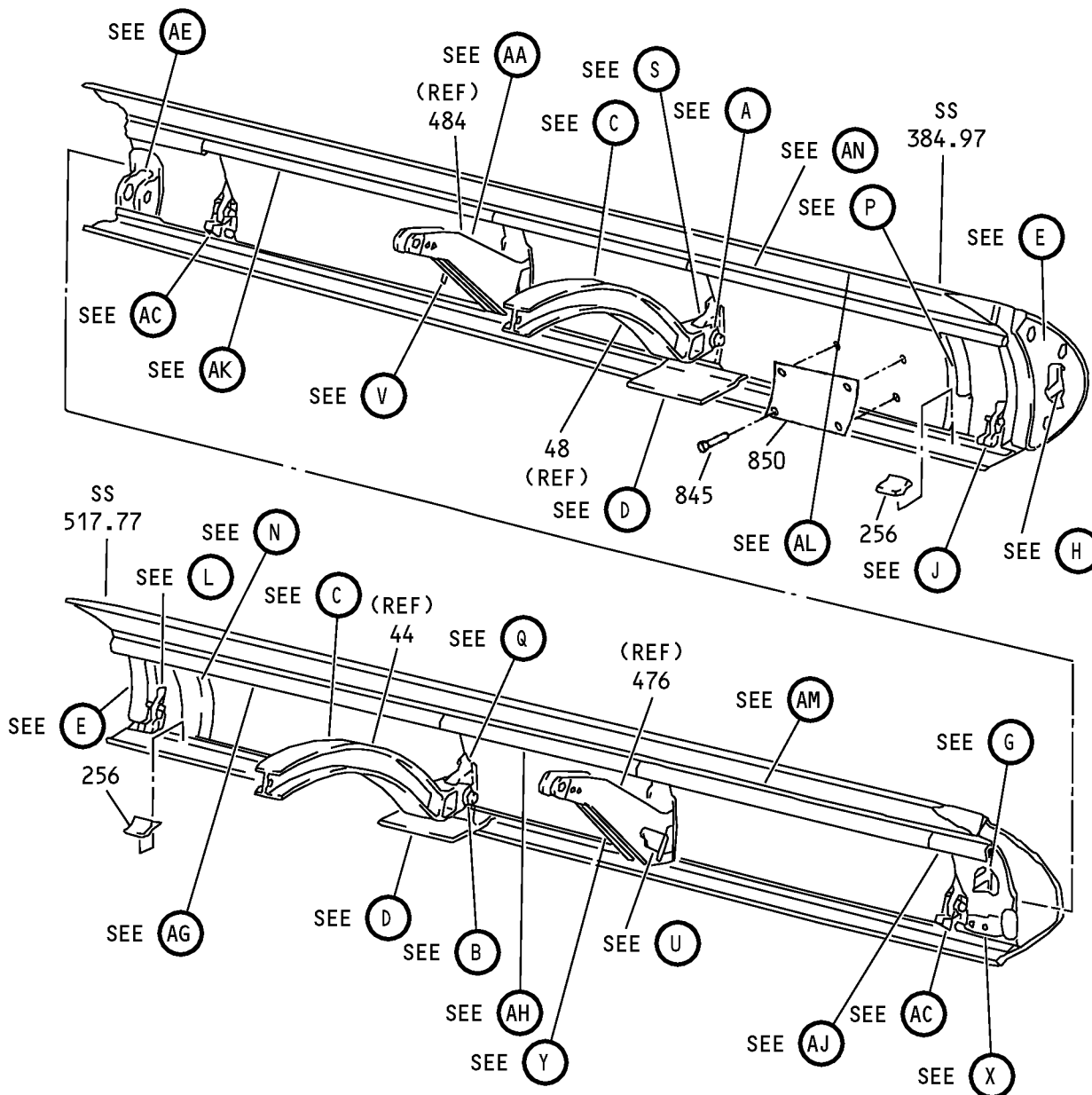
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 1 of 25)

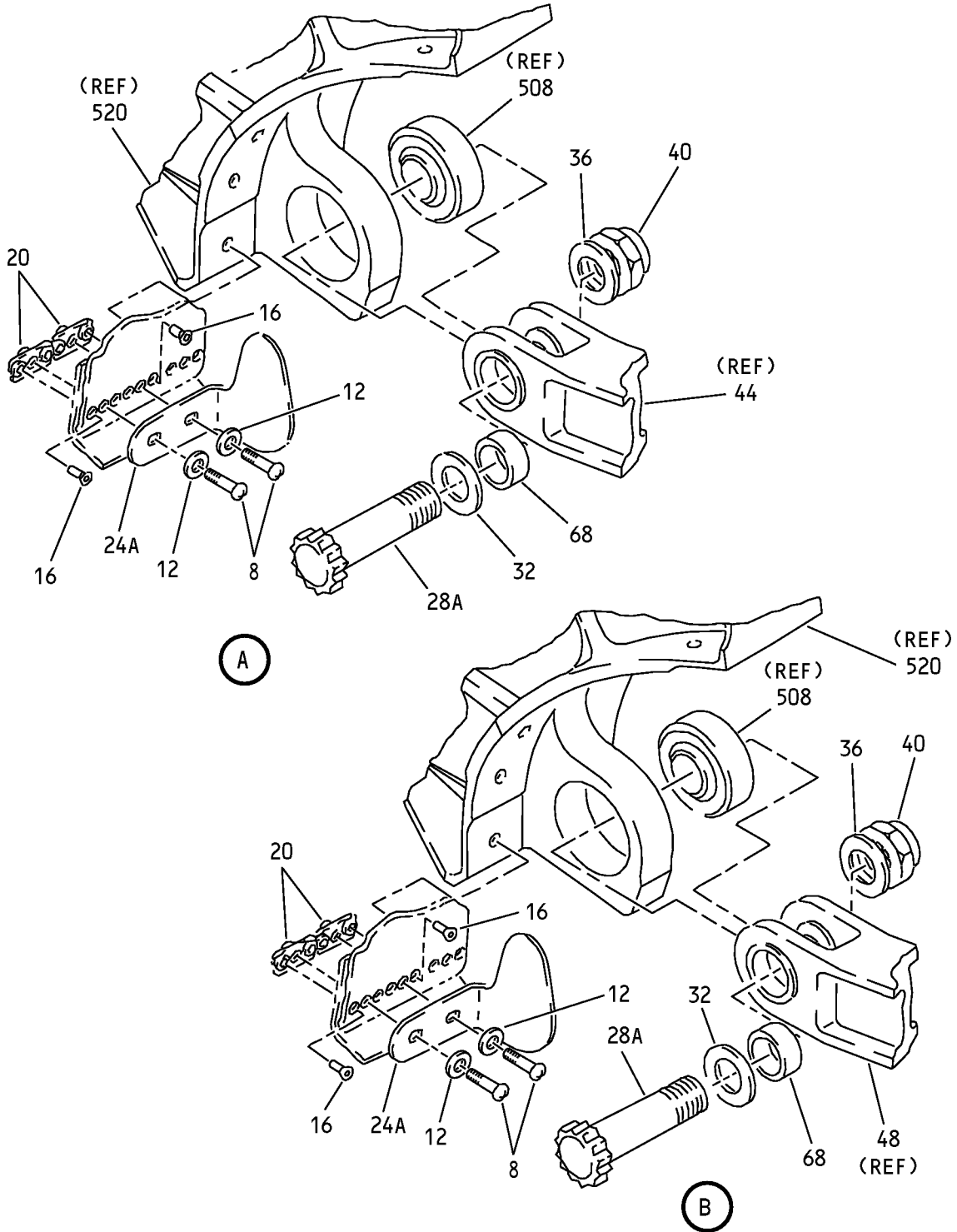
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 2 of 25)

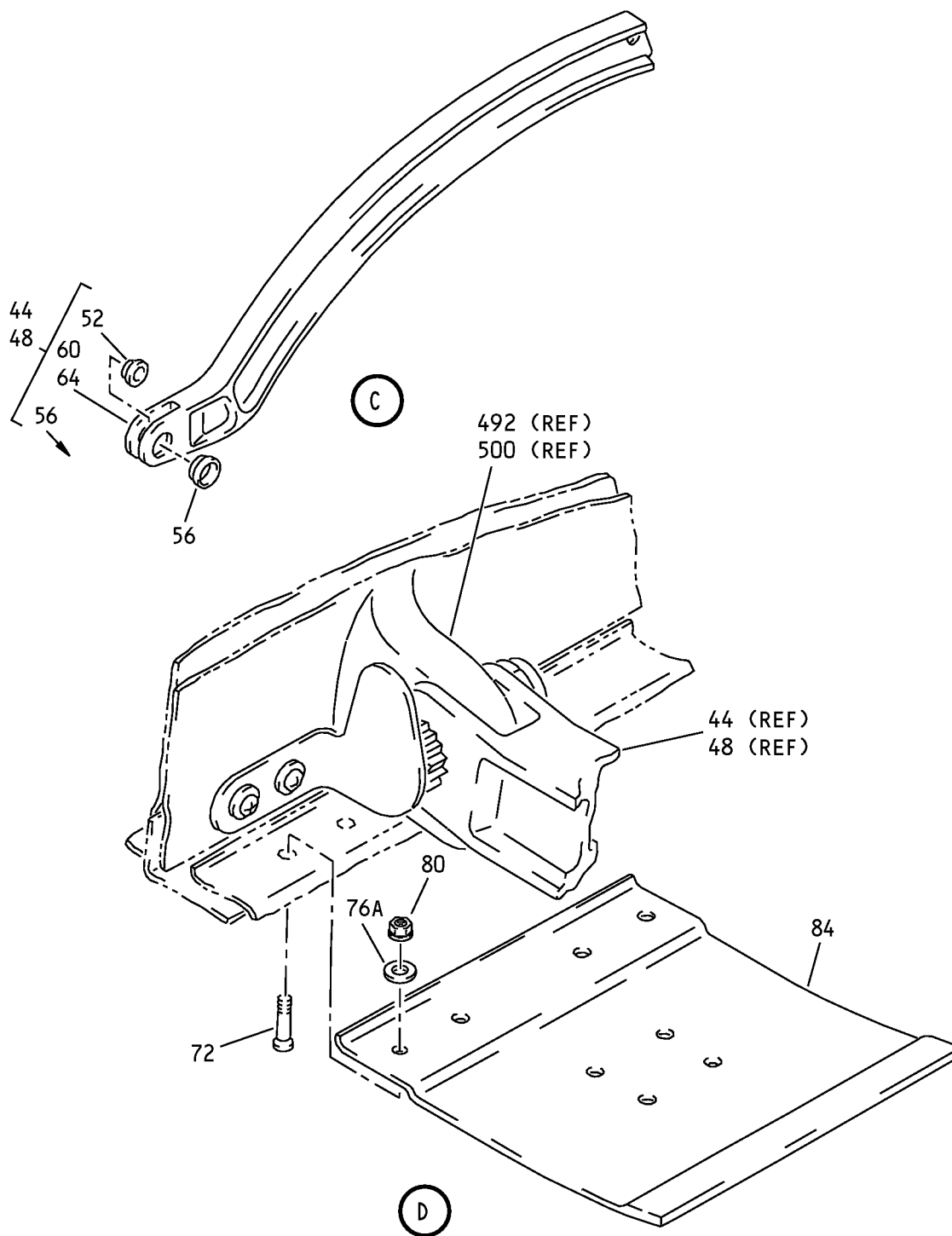
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 3 of 25)

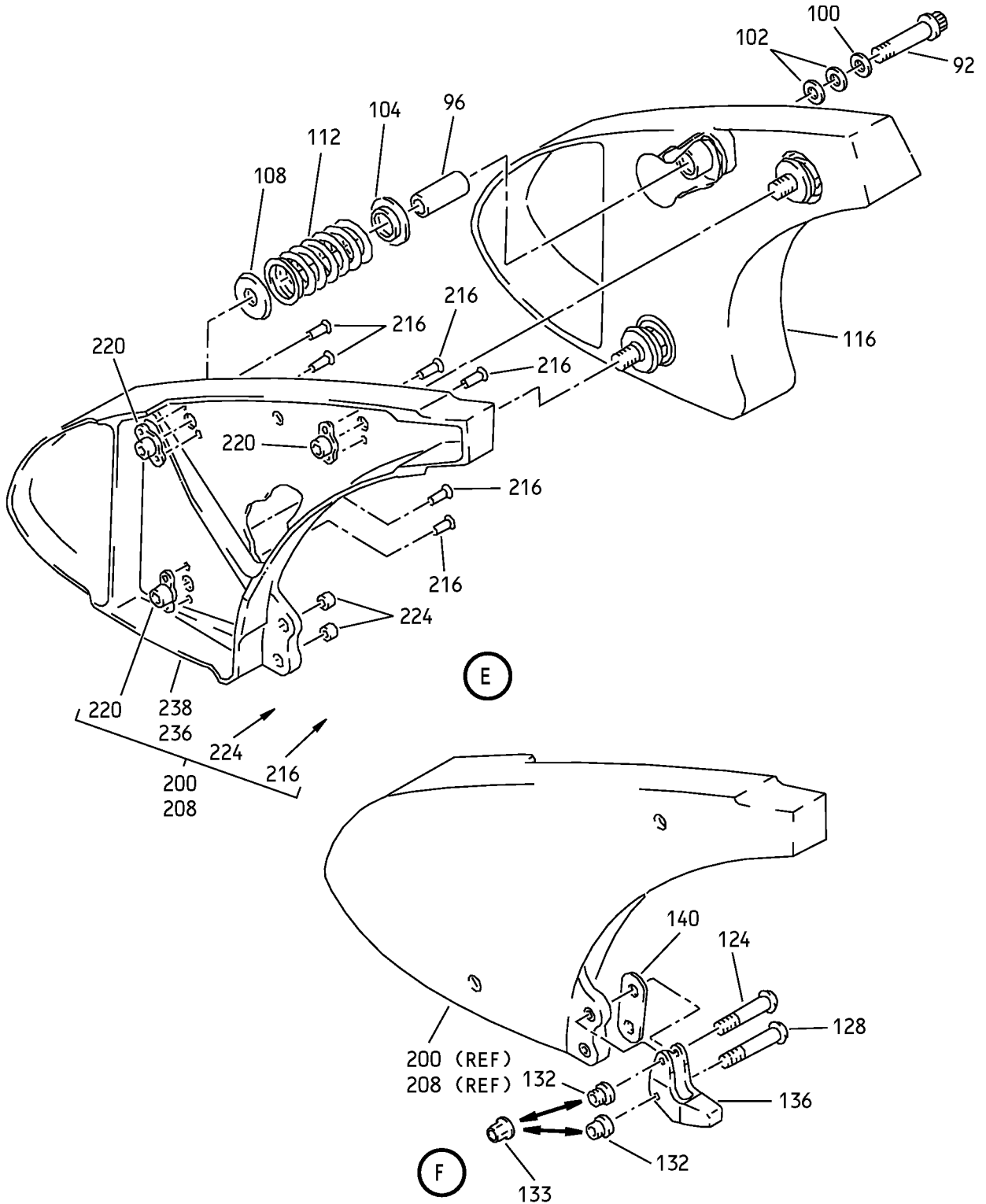
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 4 of 25)

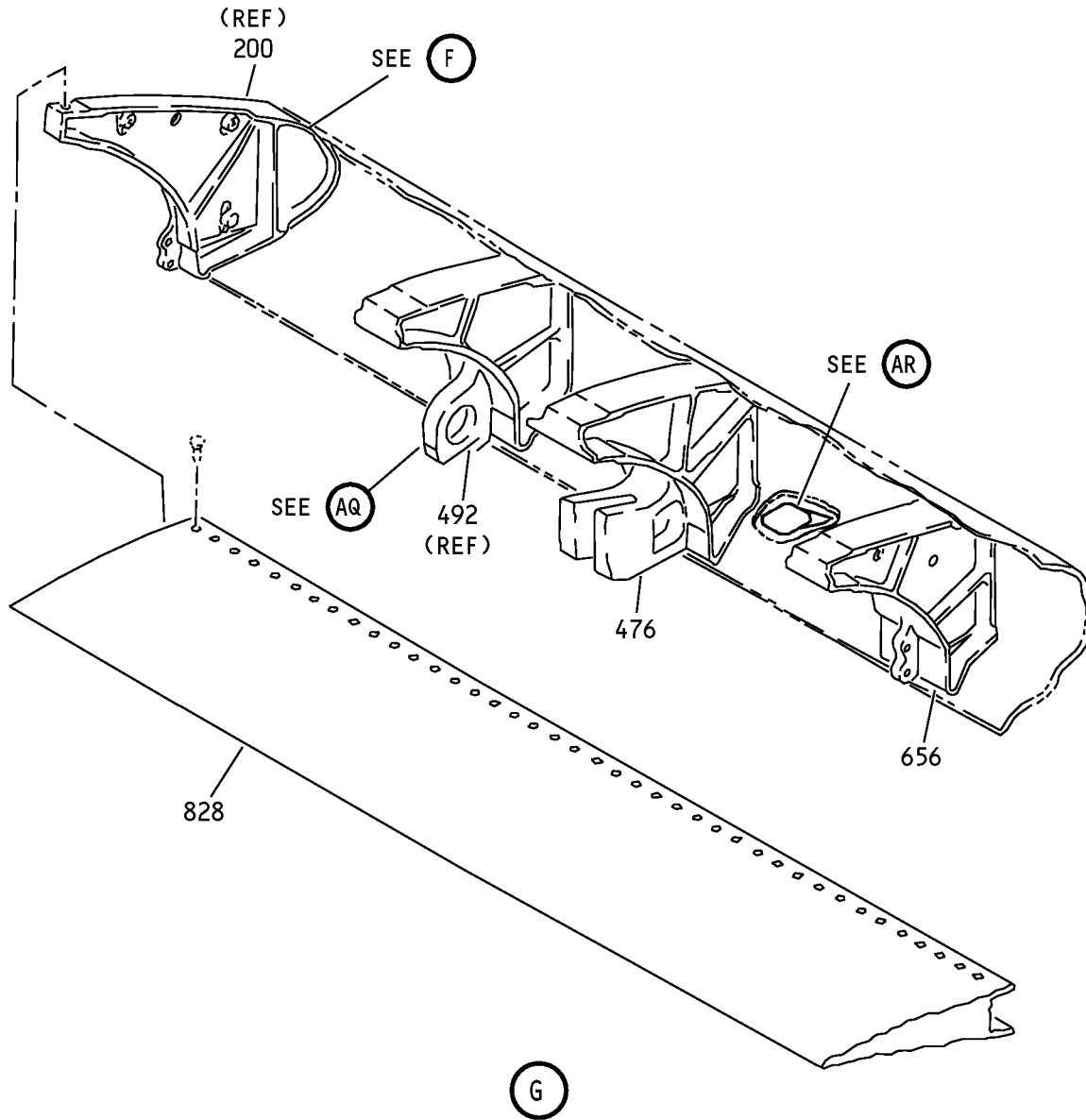
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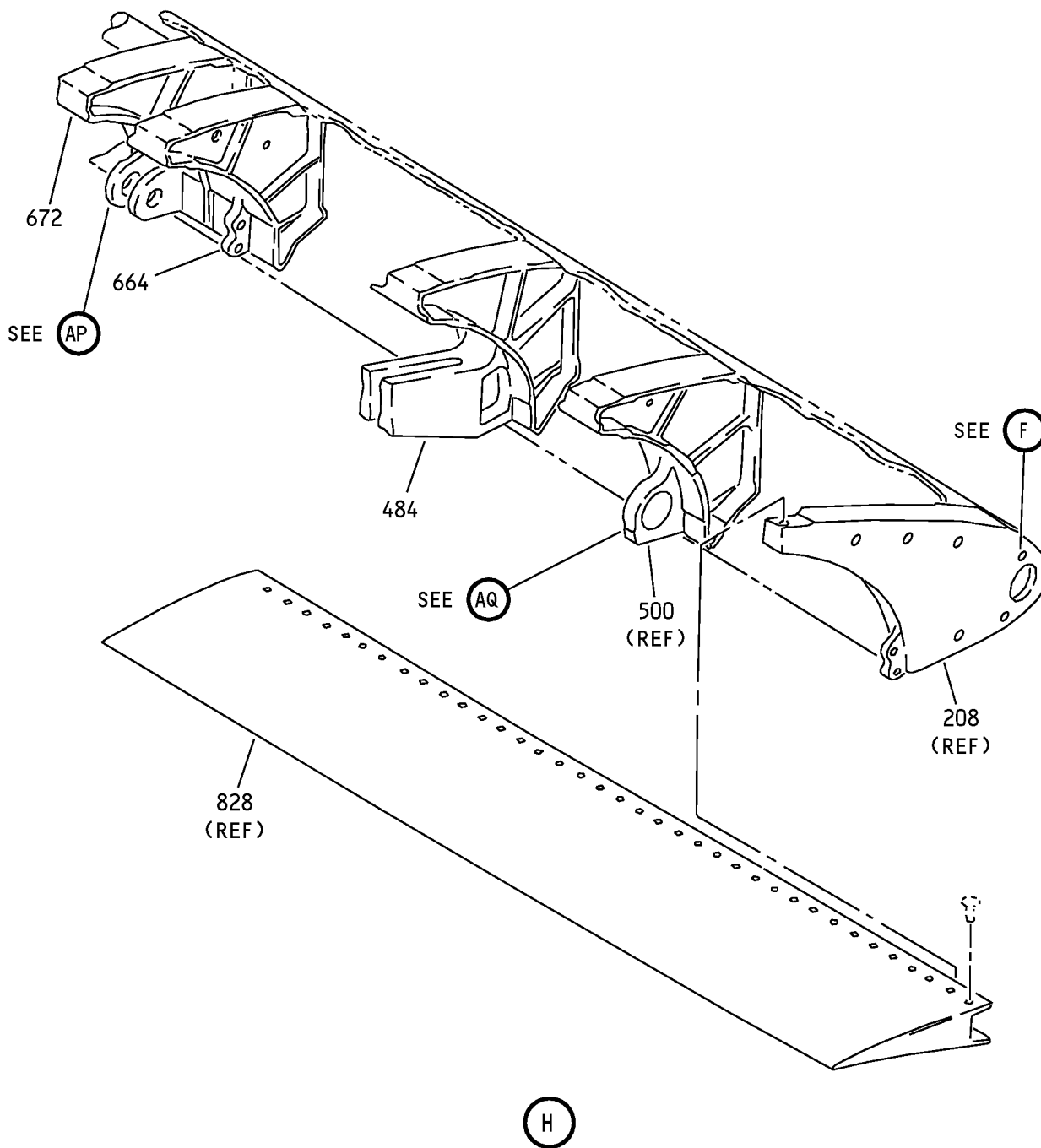
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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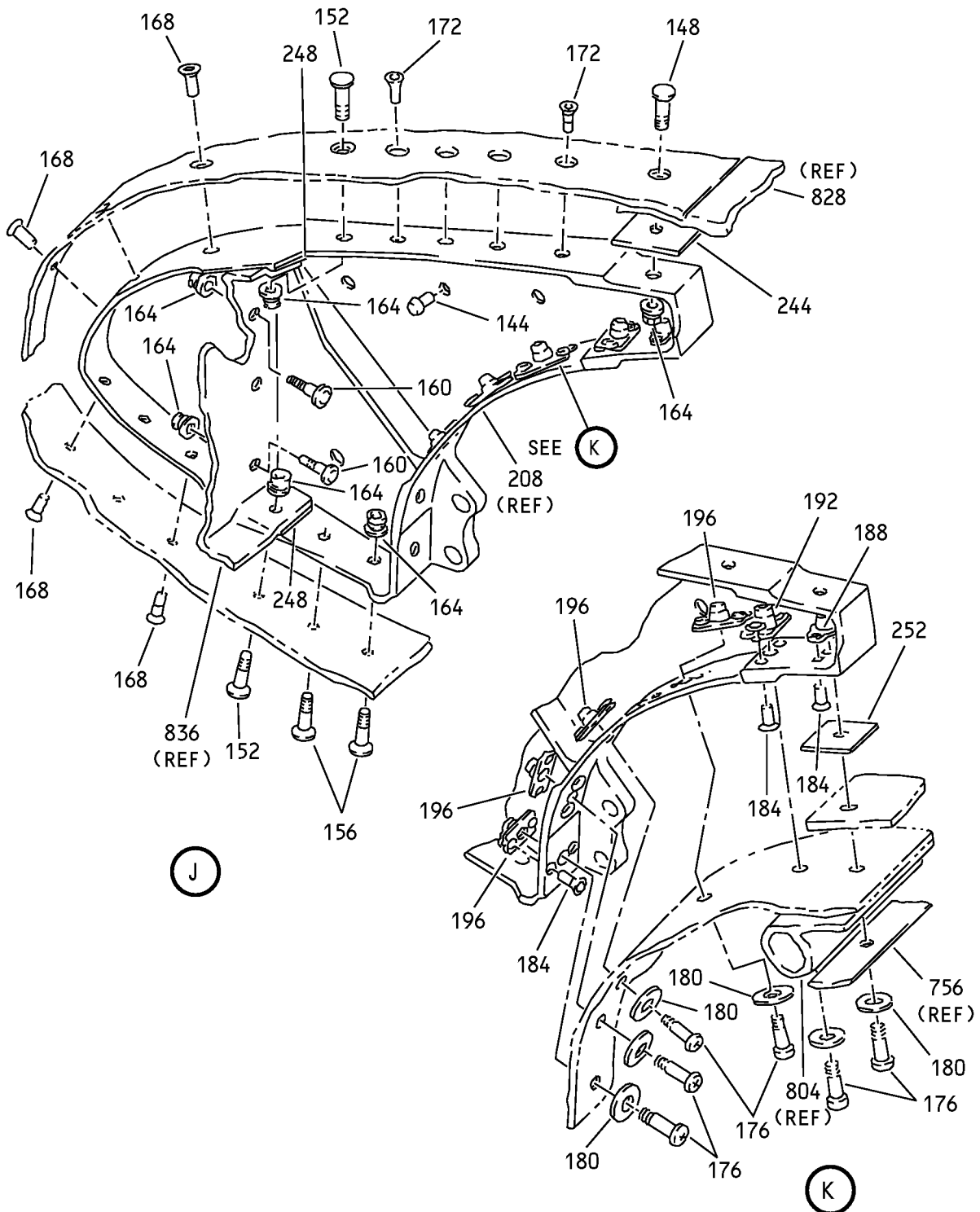
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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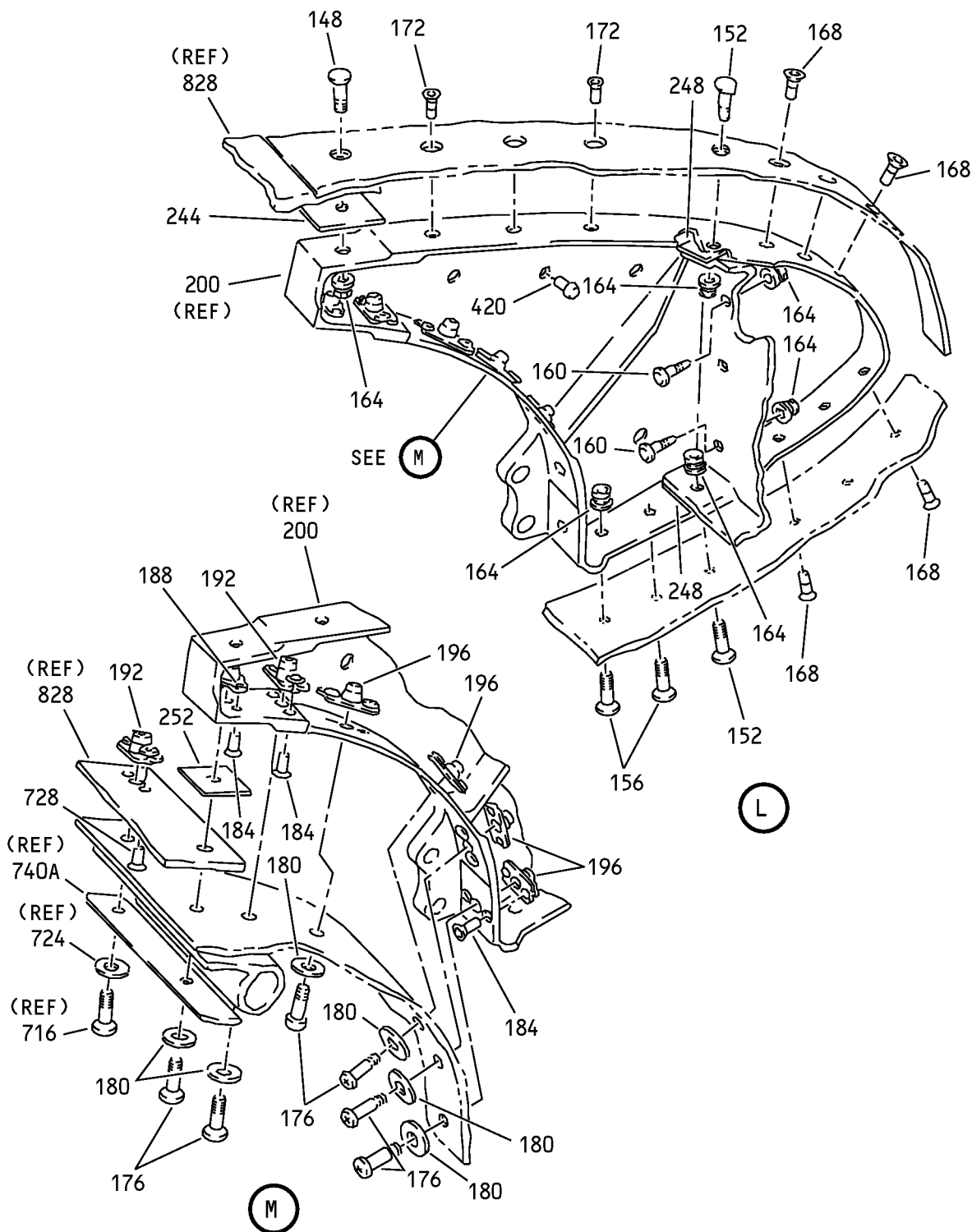
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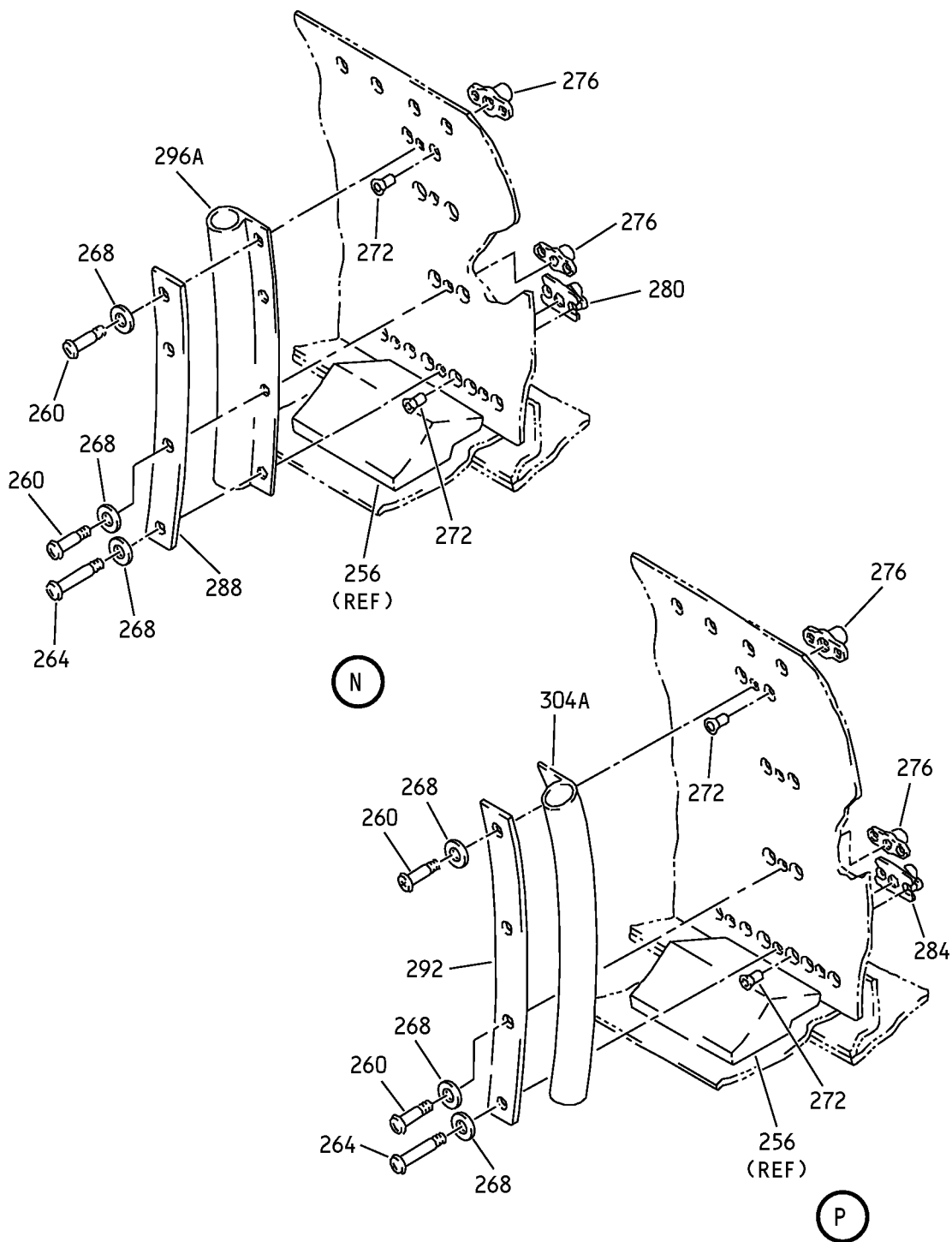
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 9 of 25)

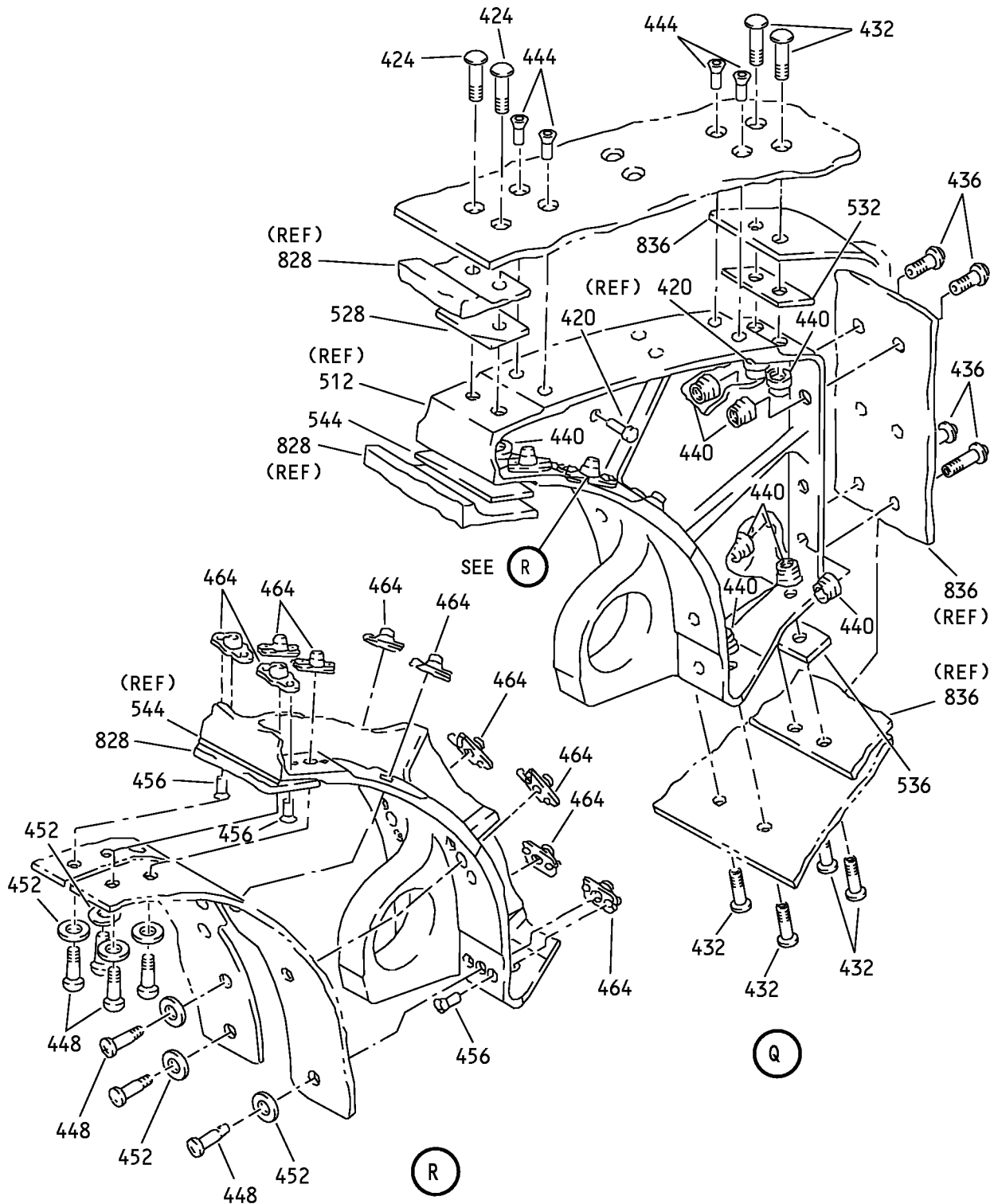
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 10 of 25)

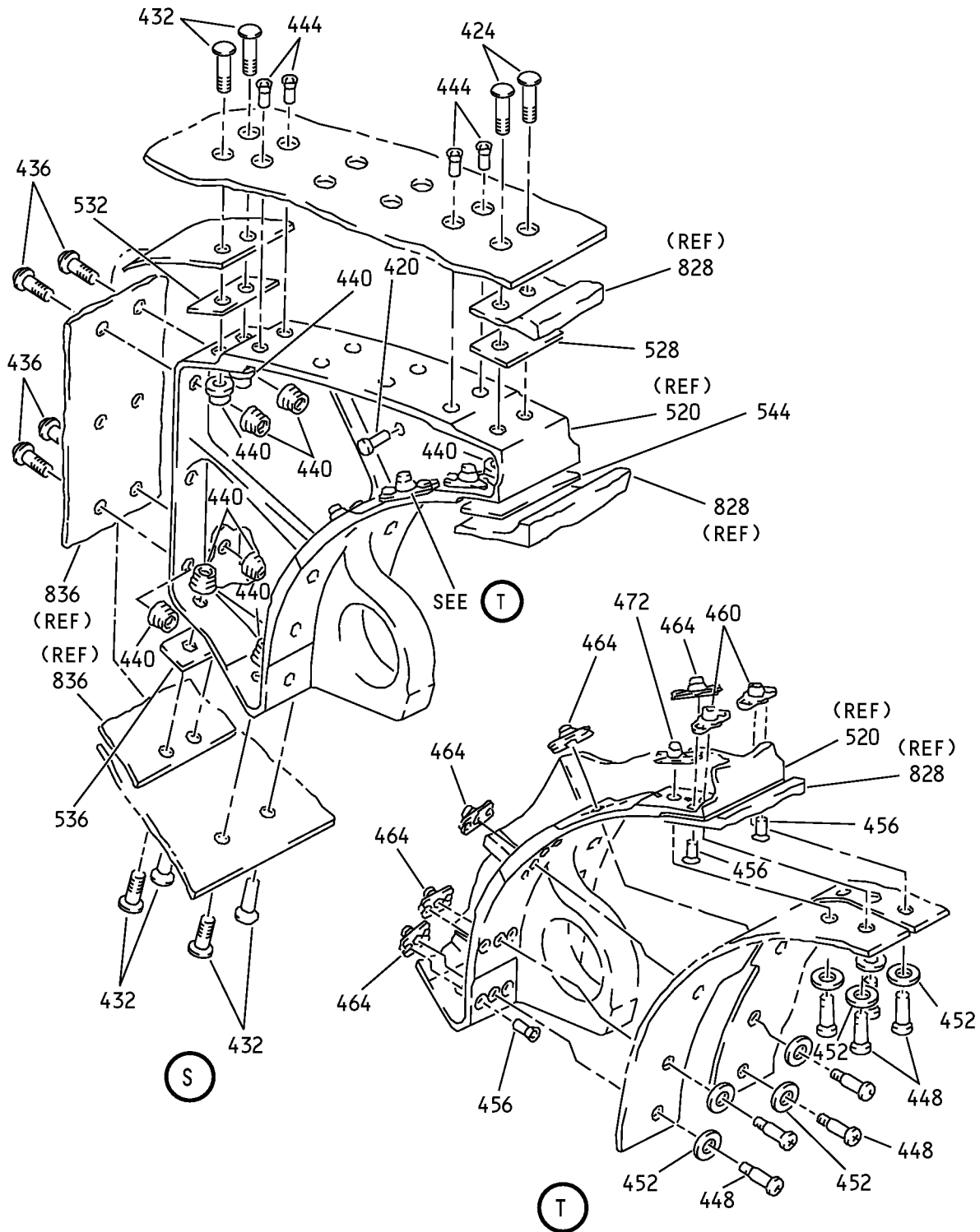
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Wing Leading Edge No. 1 and 8 Slat Assemblies
 IPL Figure 1 (Sheet 11 of 25)

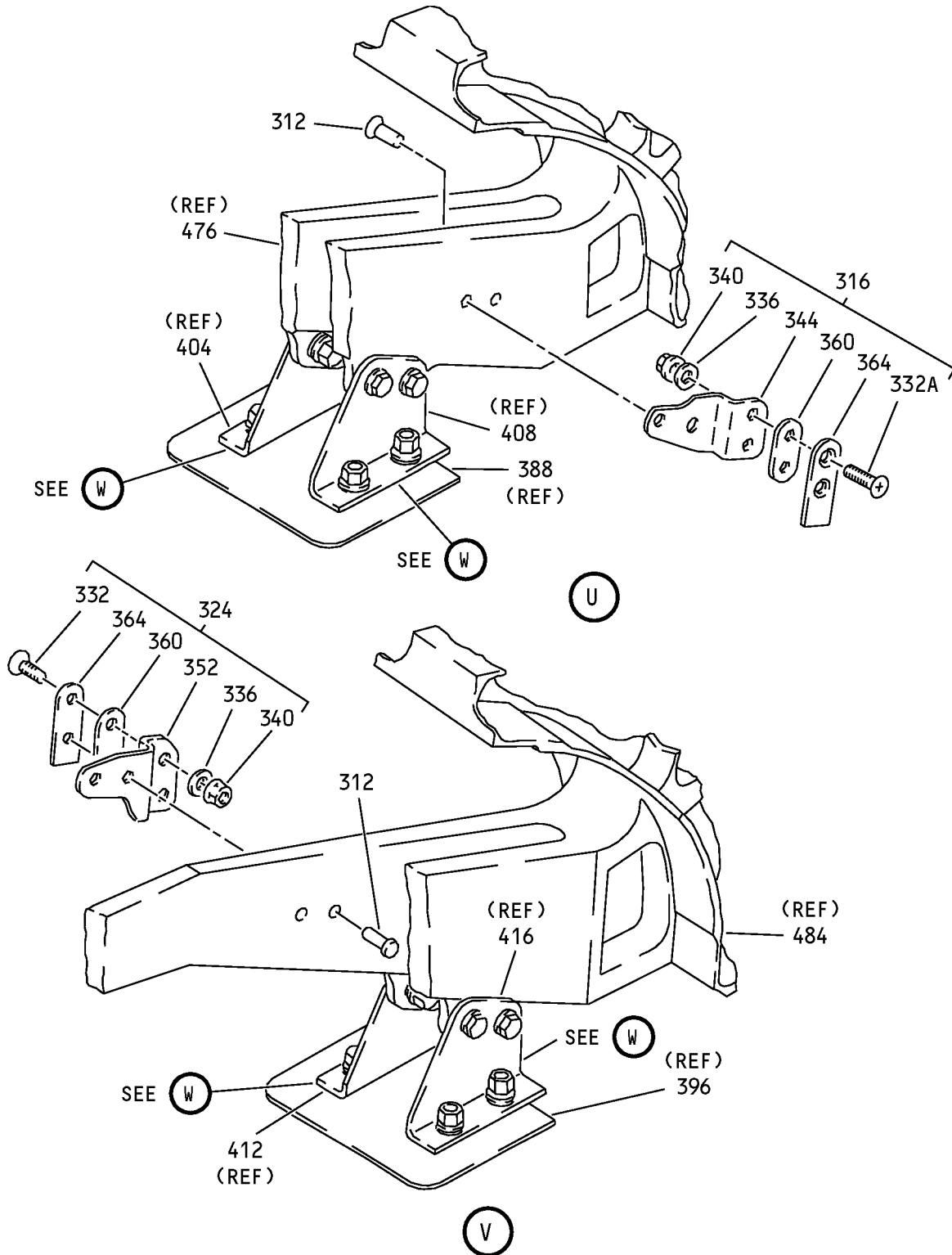
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 12 of 25)

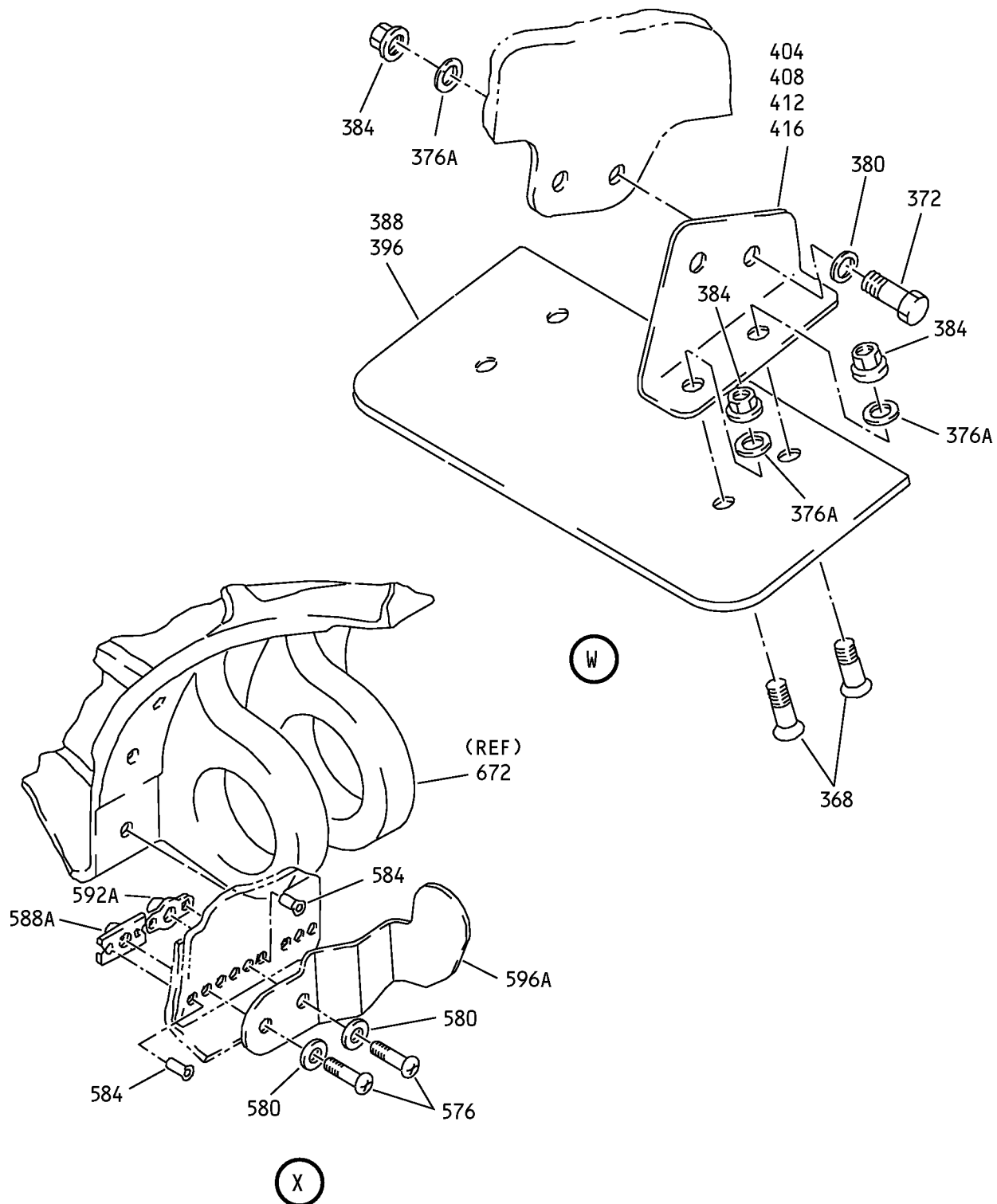
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 13 of 25)

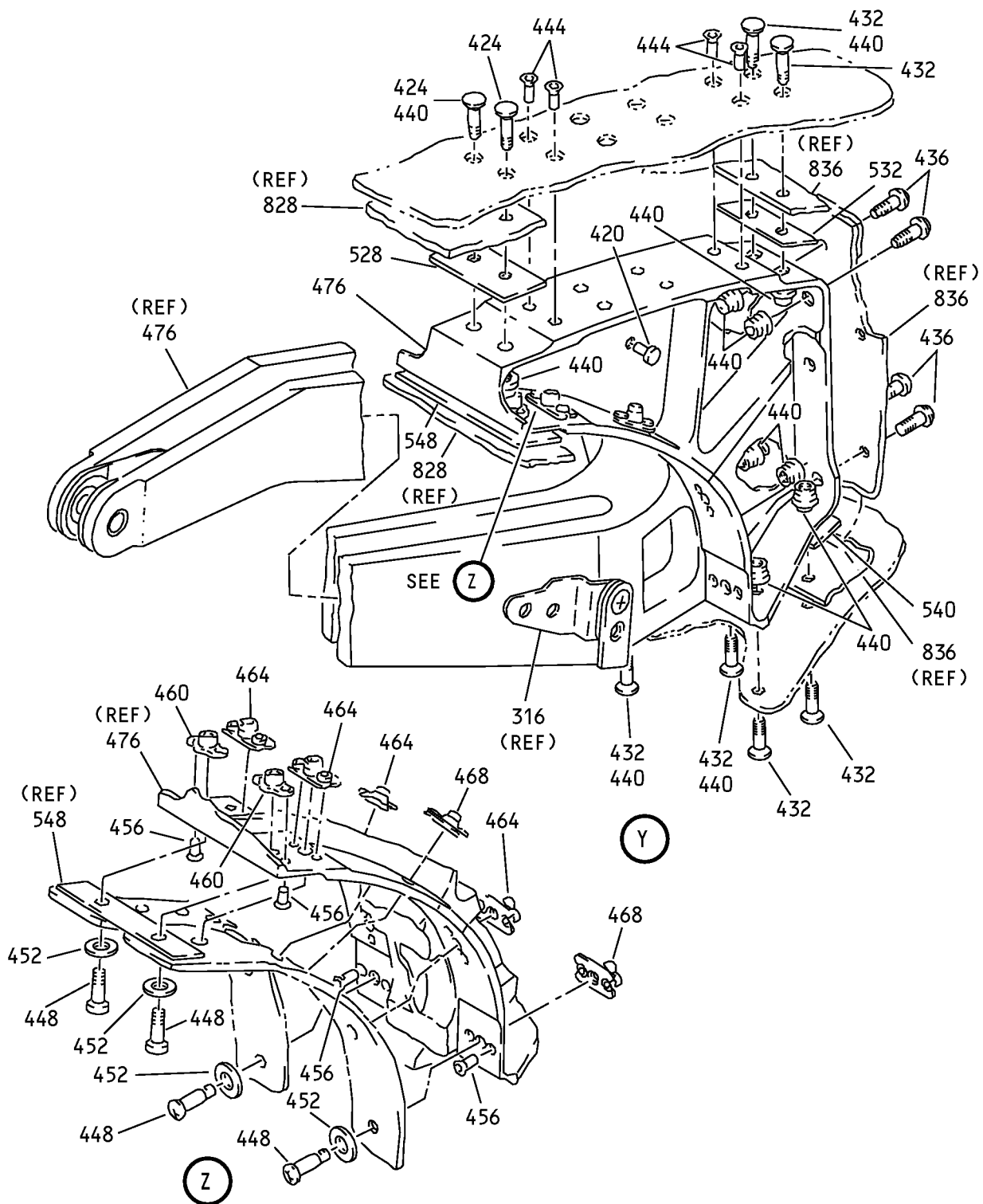
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 14 of 25)

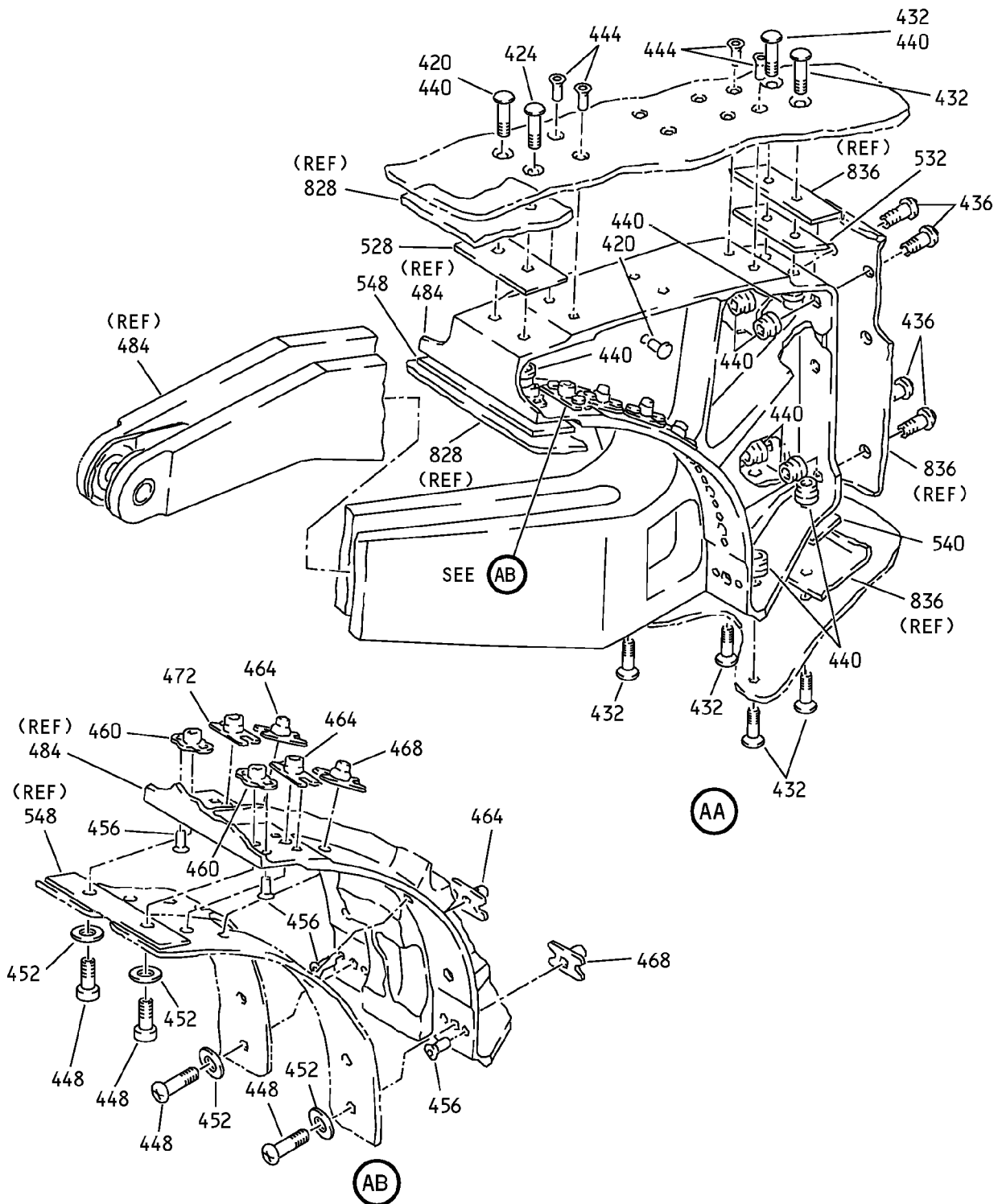
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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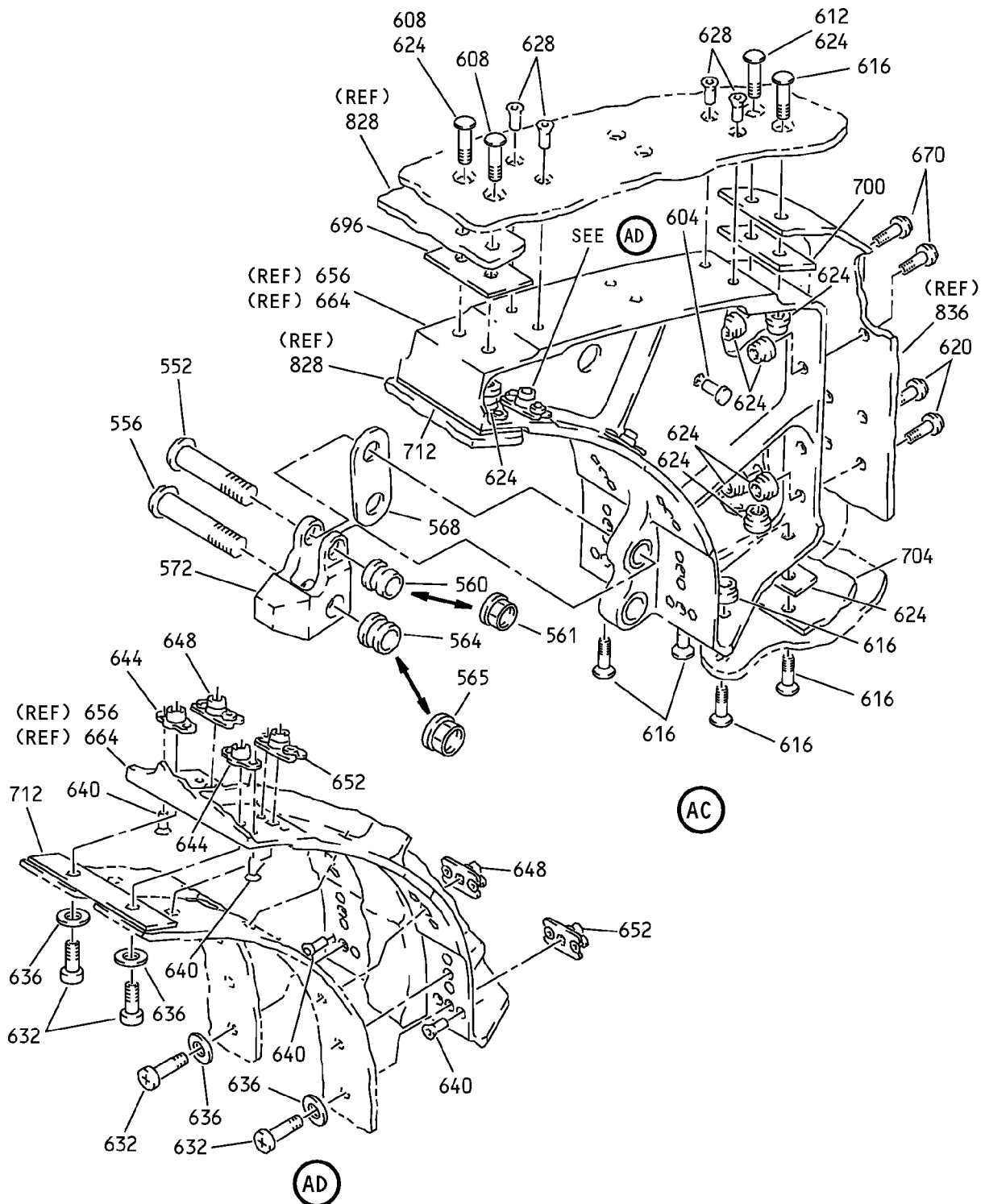
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 16 of 25)

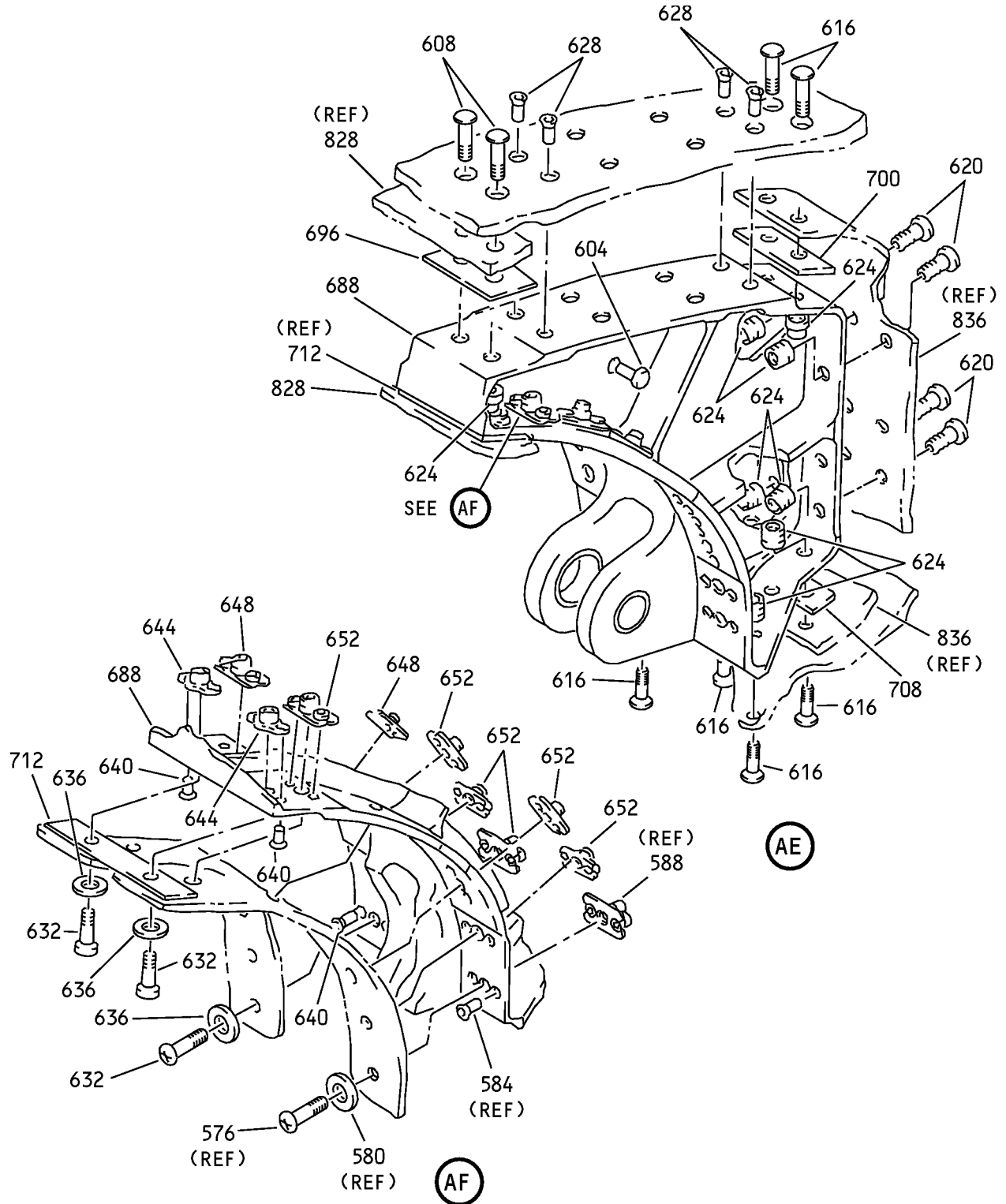
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 17 of 25)

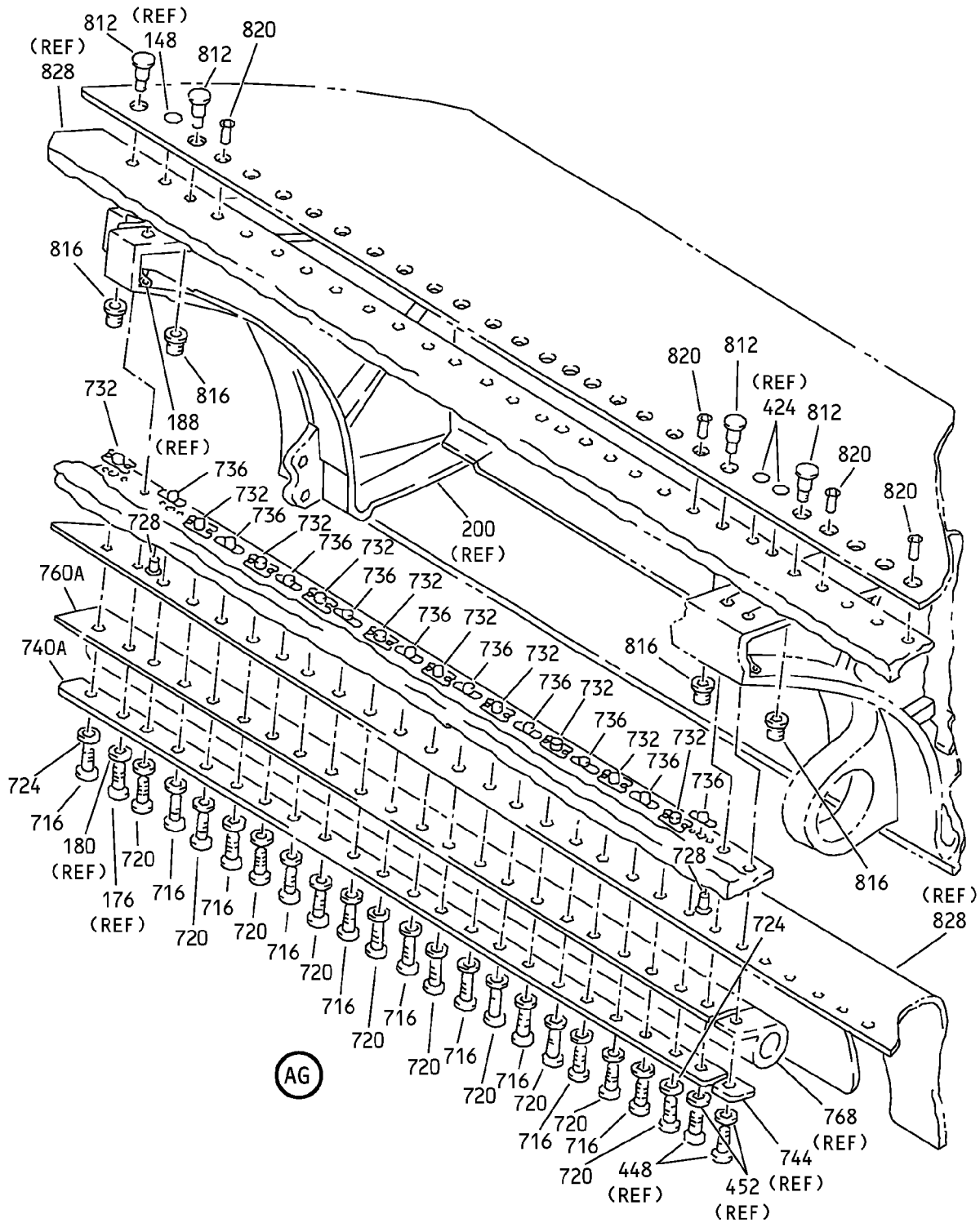
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 18 of 25)

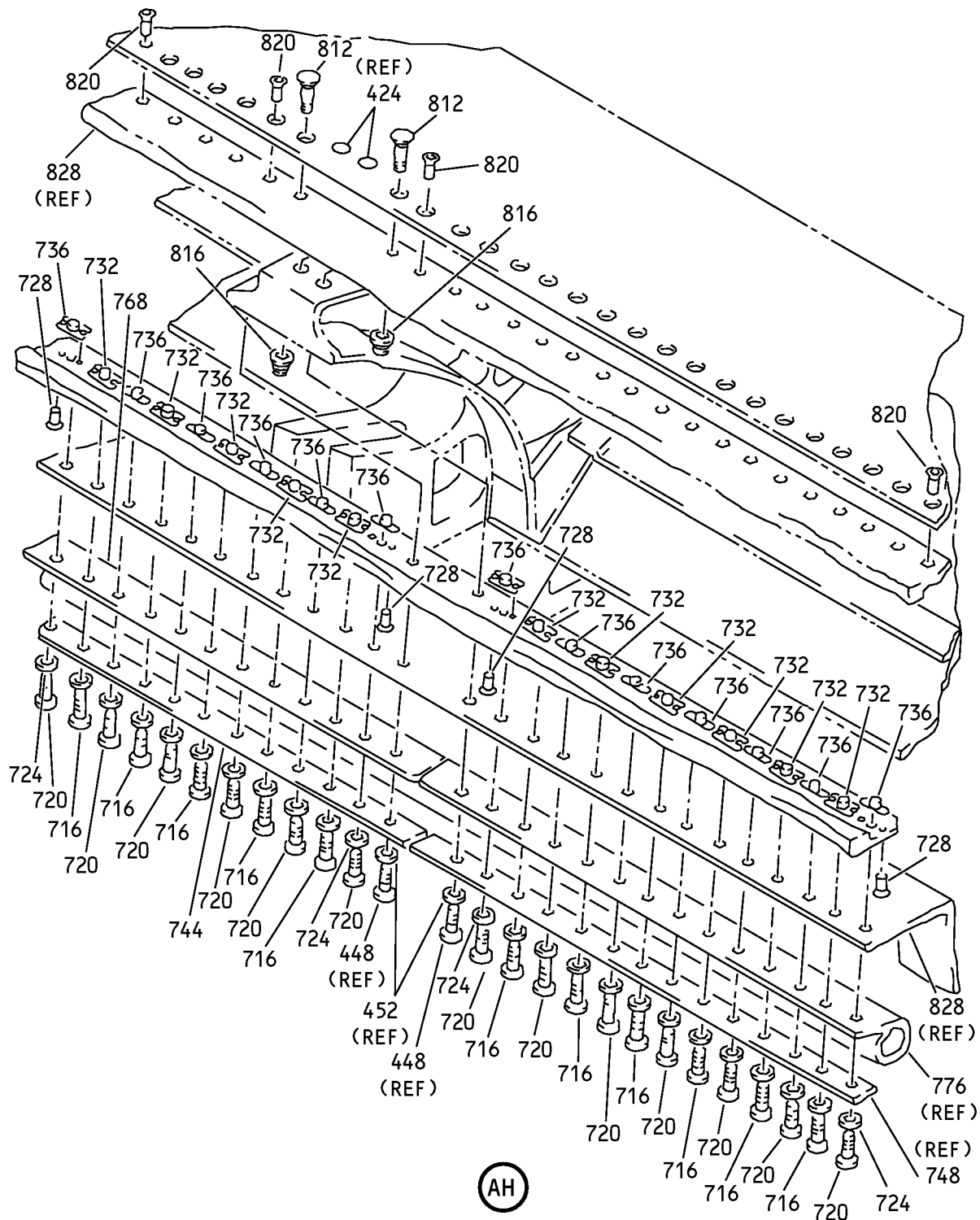
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 19 of 25)

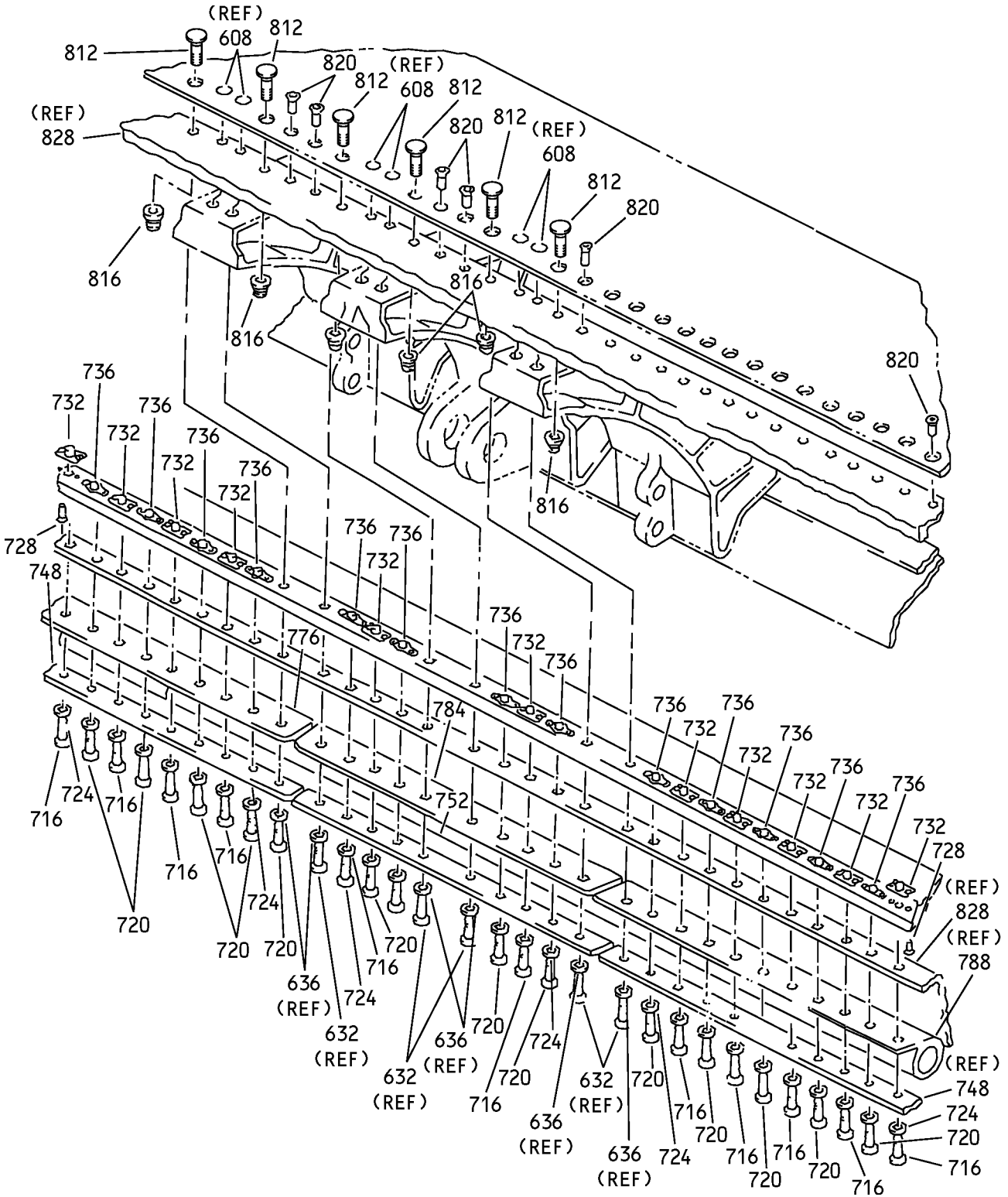
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 20 of 25)

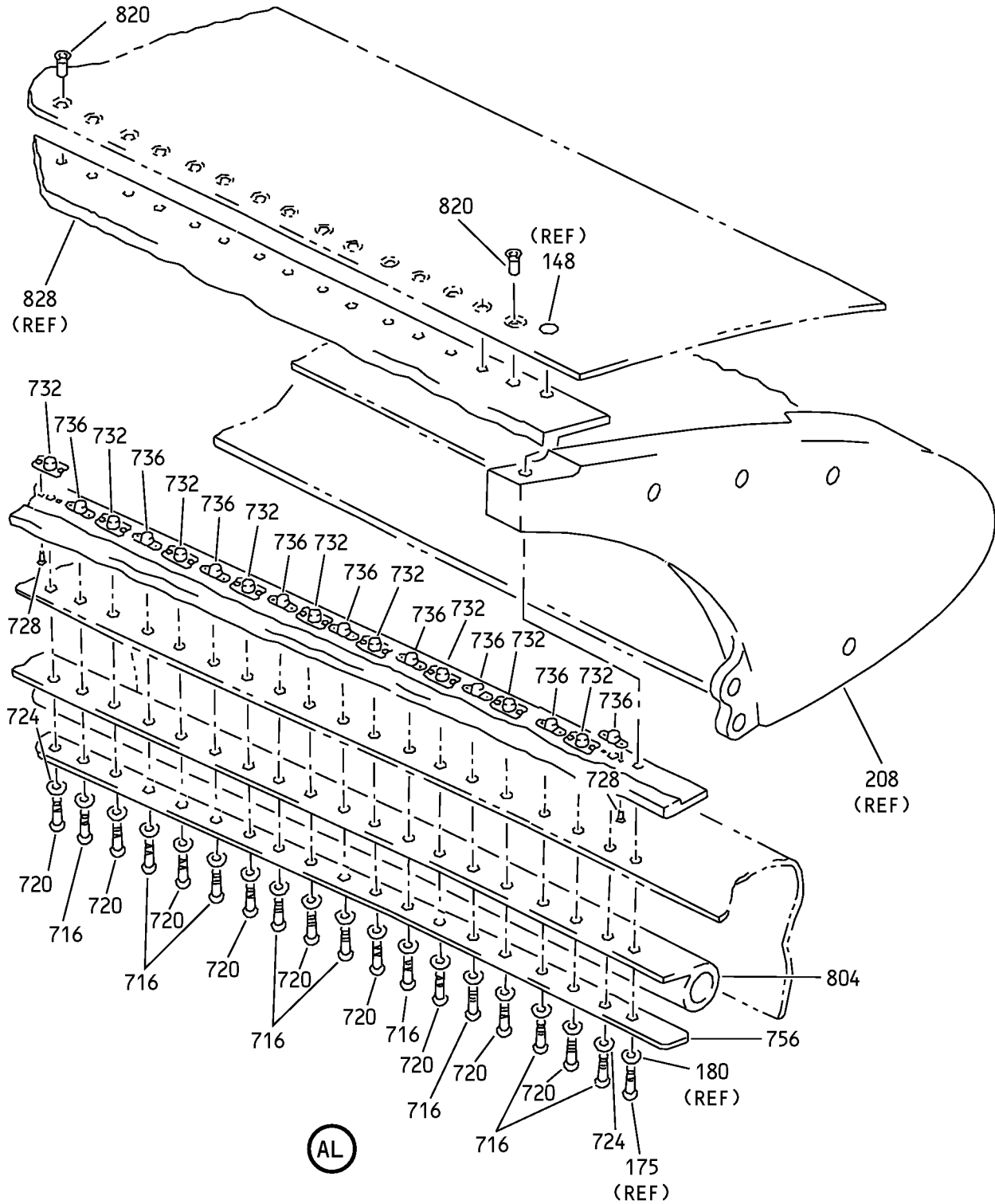
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 22 of 25)

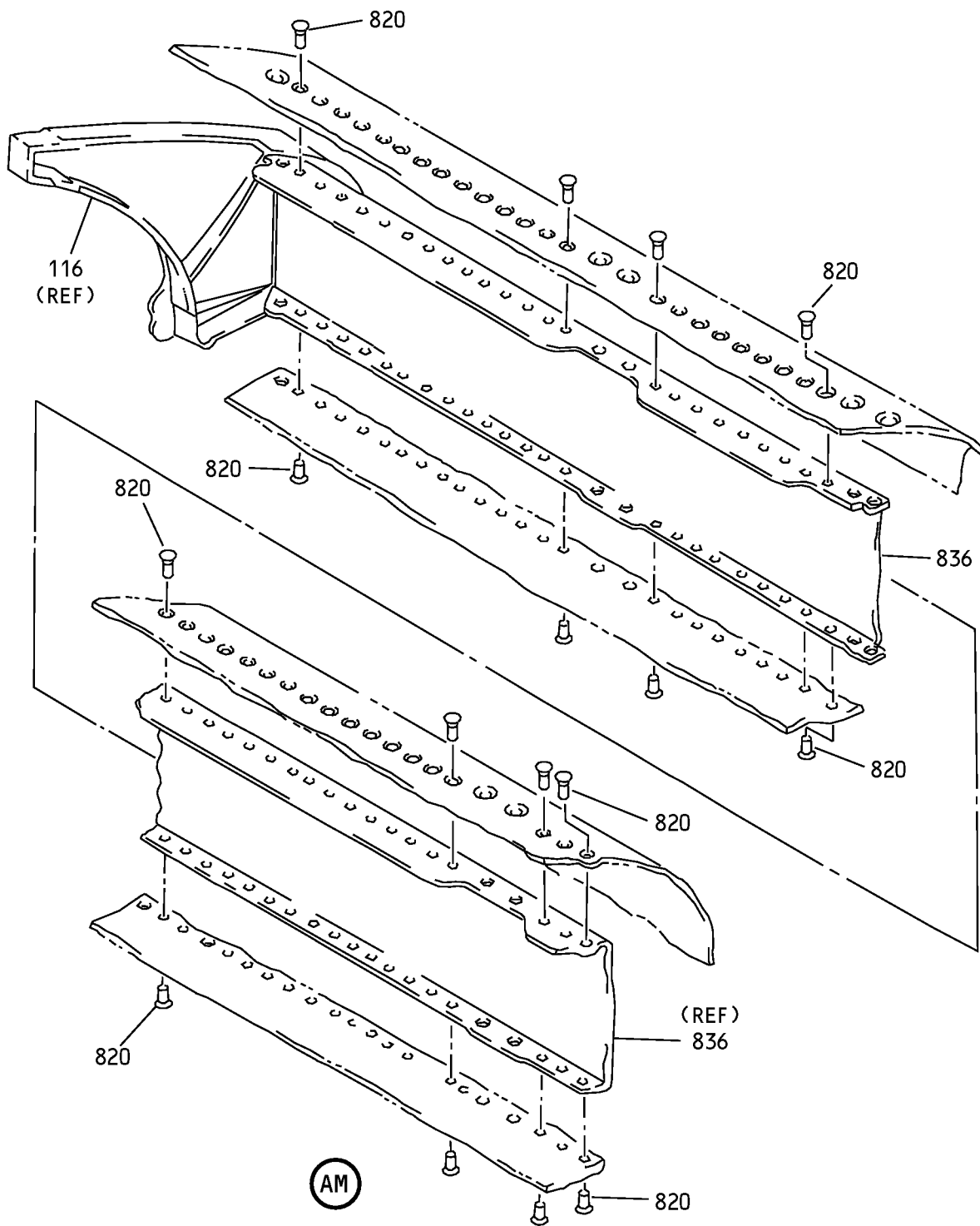
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 23 of 25)

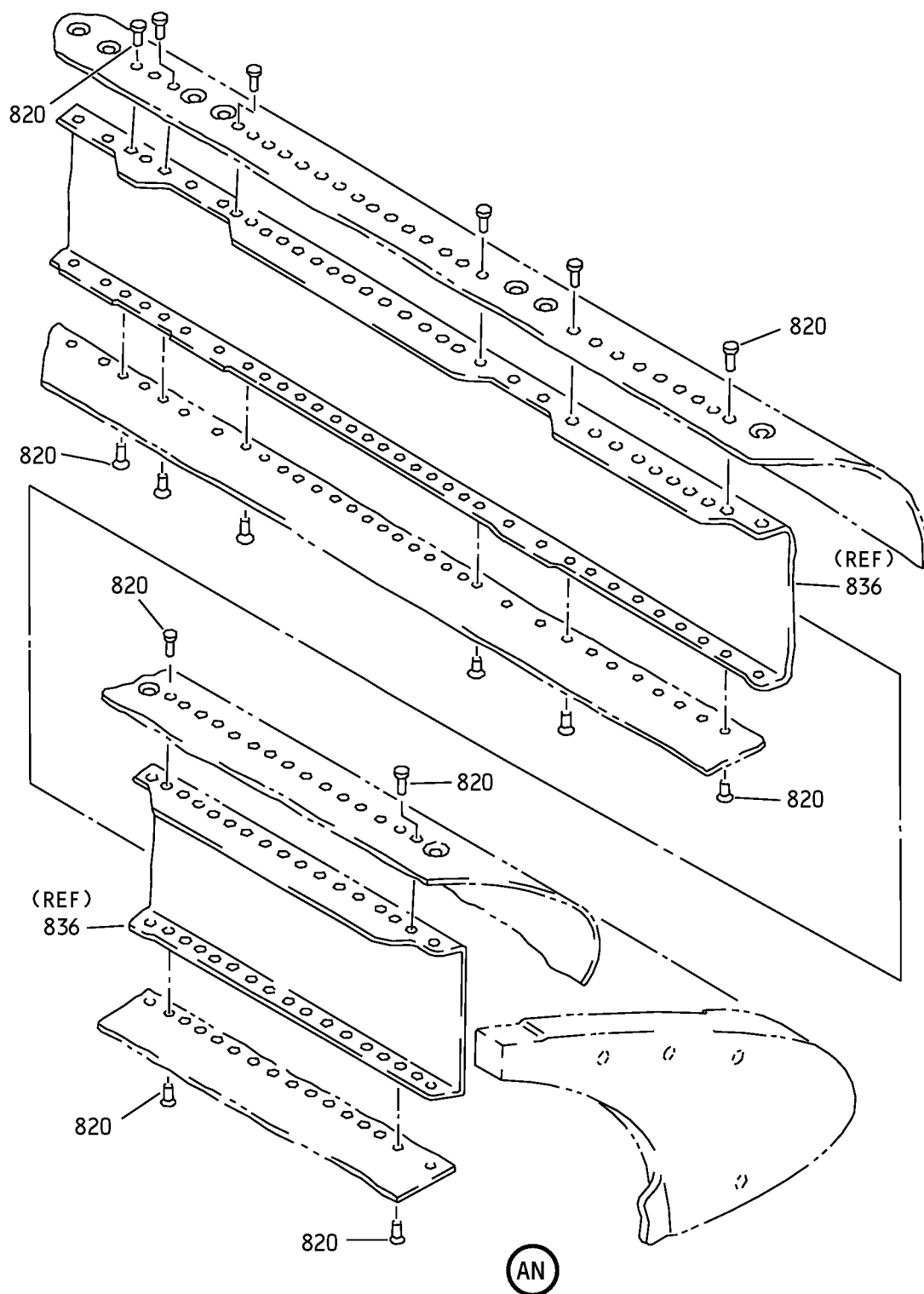
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 24 of 25)

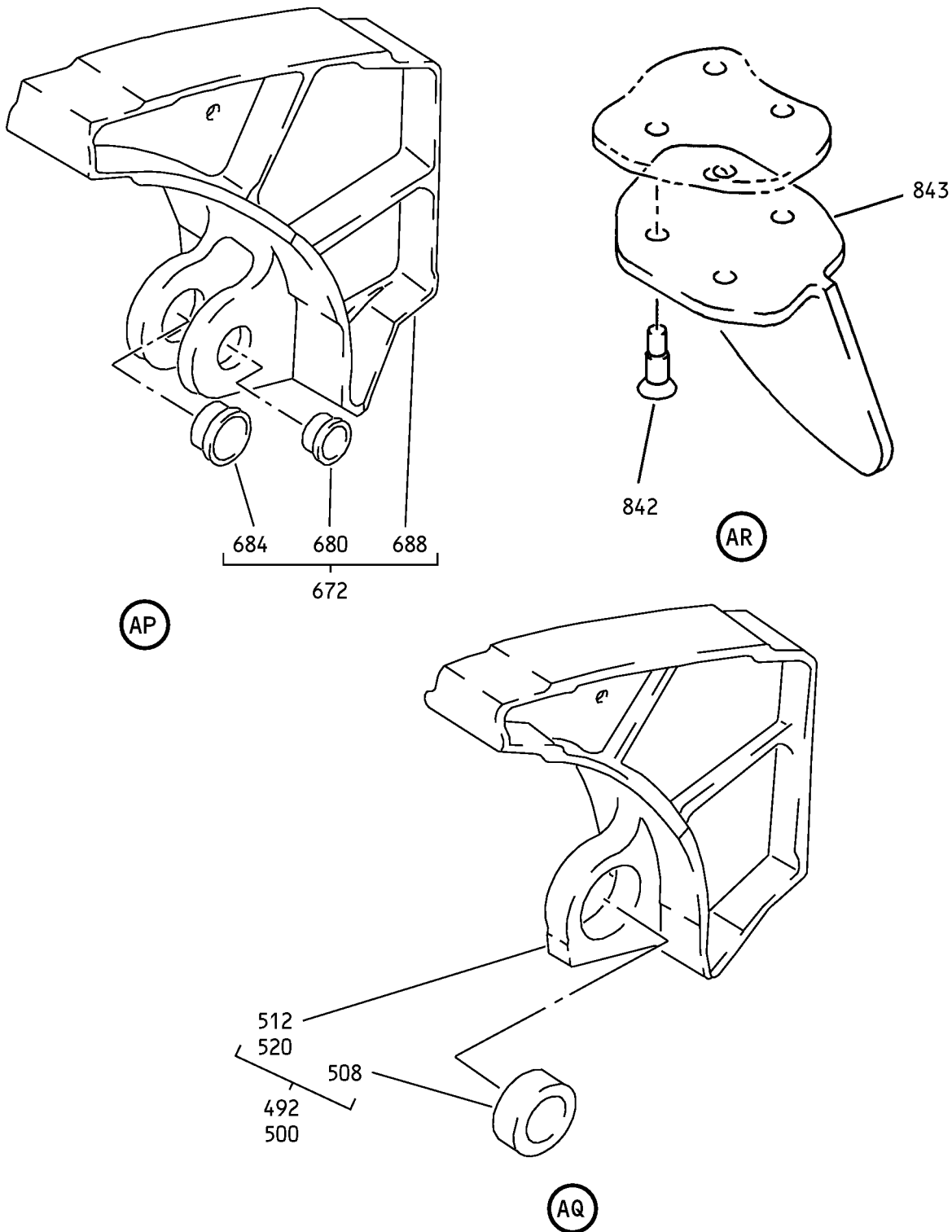
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 1 (Sheet 25 of 25)



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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-1A	114A5010-1									A	RF
-1B	114A5010-9									E	RF
-2	114A5010-3									C	RF
-2A	114A5010-201									G	RF
-2B	114A5010-11									J	RF
-4	114A5010-2									B	RF
-4A	114A5010-10									F	RF
-5	114A5010-4									D	RF
-5A	114A5010-202									H	RF
-5B	114A5010-12									K	RF
8	BACB30NT3K7									A, B, E, F	4
12	NAS1149D0316J									A, B, E, F	4
16	BACR15GE3CW									A, B, E, F	8
-16A	AF5141-3C									A, B, E, F	8
20	MF51594-3-2BAC									A, B, E, F	4
-24	114A5102-1										
24A	114A5102-11									A, B, E, F	2
-28	BACB30LT8U22										
28A	BACB30NR8K22									A, B, E, F	2
32	BACW10BP8ACU									A, B, E, F	2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
36	BACW10BP8APU		.	W	A	S	H	E	R	A, B, E, F	2
40	H01-8BAC		.	N						A, B, E, F	2
				(V15653)							
				(SPEC BACN10JC8CM)							
				(OPT BMN4122C1D2-8 (V85495))							
				(OPT 109LH9074-8 (V72962))							
				(OPT 69235-820CM (V56878))							
				(OPT BMN4122C1D2-8 (V97928))							
44	114A7511-1		.	T	R	A	C	K	A	S	S
				(OPT ITEM 44A)						A, B, E, F	1
-44A	114A7511-5		.	T	R	A	C	K	A	S	S
				(OPT ITEM 44)						A, B, E, F	1
48	114A7512-1		.	T	R	A	C	K	A	S	S
				(OPT ITEM 48A)						A, B, E, F	1
-48A	114A7512-5		.	T	R	A	C	K	A	S	S
				(OPT ITEM 48)						A, B, E, F	1
52	BACB28AP08P018		.	B						A, B, E, F	1
56	BACB28AT11B018C		.	B						A, B, E, F	1
60	114A7511-2		.	T						A, B, E, F	1
				(USED ON ITEM 44)							
-60A	114A7511-6		.	T						A, B, E, F	1
				(USED ON ITEM 44A)							
64	114A7512-2		.	T						A, B, E, F	1
				(USED ON ITEM 48)							
-64A	114A7512-6		.	T						A, B, E, F	1
				(USED ON ITEM 48A)							
68	BACB28AK08-034		.	B						A, B, E, F	2
72	HST11AG6-5		.	B						A, B, E, F	8
				(V06725)							
				(SPEC BACB30VU6K5)							
				(OPT HST11AG6-5 (V73197))							
				(OPT HST11AG6-5 (V56878))							
				(OPT HST11AG6-5 (V0PTK6))							
-76	NAS1149D0332J			DELETED							
76A	NAS1149D0363J		.	W						A, B, E, F	8
80	H52732-3CD		.	N						A, B, E, F	8
				(V15653)							
				(SPEC BACN10YR3CD)							
				(OPT PLH53CD (V62554))							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
84	114A9202-1		.	TAB-FLEX						A, E	2
-88	114A9202-2		.	TAB-FLEX						B, F	2
92	BACB30MR4K16		.	BOLT						A, B, E, F	3
96	NAS42DD8A54FC		.	SPACER						A, B, E, F	3
100	69-77559-1		.	WASHER-ALUMINUM						A, B, E, F	3
102	69-77559-3		.	WASHER-ALUMINUM						A, B, E, F	2
104	69-77600-1		.	SLEEVE						A, B, E, F	3
108	69-77562-1		.	RETAINER-SPR						A, B, E, F	3
112	114W4709-1		.	SPRING						A, B, E, F	3
116	114A9011-1		.	SEAL-END, OUTBD						A, E	1
-120	114A9011-2		.	SEAL-END, OUTBD						B, F	1
124	HLT422AP6-10		.	BOLT (V97928) (SPEC BACB30YK6K10) (OPT HLT422AP6-10 (V06725)) (OPT HLT422AP6-10 (V60516)) (OPT ITEM 124A)						A, B, E, F	2
-124A	HLT422SE6-10		.	BOLT (V73197) (OPT ITEM 124)						A, B, E, F	2
128	HLT422AP6-11		.	BOLT (V97928) (SPEC BACB30YK6K11) (OPT HLT422AP6-11 (V06725)) (OPT HLT422AP6-11 (V60516)) (OPT ITEM 128A)						A, B, E, F	2
-128A	HLT422PB6-11		.	BOLT (V73197) (OPT ITEM 128)						A, B, E, F	2
132	BACC30CP6S		.	COLLAR (OPT ITEM 132A, 133)						A, B, E, F	4

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE	USAGE CODE	UNITS PER ASSY
1- -132A	HL1187DU6		. COLLAR (V5M902) (SPEC BACC30X6S) (OPT HL87DU6 (V92215)) (OPT HL87DU6 (V73197)) (OPT HL1187DU6 (V56878)) (OPT HL1187DU6 (V92215)) (OPT HL87DU6 (V56878)) (OPT HL1187DU6 (V73197)) (OPT ITEM 132, 133)	A, B, E, F	4
133	BACN10YT3CS		. NUT (OPT ITEM 132, 132A)	A, B, E, F	4
136	114A7701-1		. FITTING-END STOP	A, B, E, F	2
140	114A5107-1		. SHIM-END STOP	A, B, E, F	2
144	BACR15FT8D		. RIVET (SIZE DETERMINED ON INST) (OPT ITEM 144A)	A, B, E, F	2
-144A	BACR15FT8AD		. RIVET (SIZE DETERMINED ON INST) (OPT ITEM 144)	A, B, E, F	2
148	HST11AG6-5		. BOLT (V06725) (SPEC BACB30VU6K5) (OPT HST11AG6-5 (V73197)) (OPT HST11AG6-5 (V56878)) (OPT HST11AG6-5 (V0PTK6))	A, B, E, F	2
152	HST11AG6-4		. BOLT (V06725) (SPEC BACB30VU6K4) (OPT HST11AG6-4 (V73197)) (OPT HST11AG6-4 (V56878)) (OPT HST11AG6-4 (V0PTK6))	A, B, E, F	4
156	HST11AG6-3		. BOLT (V06725) (SPEC BACB30VU6K3) (OPT HST11AG6-3 (V73197)) (OPT HST11AG6-3 (V56878)) (OPT HST11AG6-3 (V0PTK6))	A, B, E, F	4

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
160	HST10AG6-3		.	BOLT						A, B, E, F	6
				(V0PTK6)							
				(SPEC BACB30VT6K3)							
				(OPT HST10AG6-3 (V06725))							
				(OPT HST10AG6-3 (V56878))							
				(OPT HST10AG6-3 (V73197))							
164	HST79CY6		.	COLLAR						A, B, E, F	16
				(V73197)							
				(SPEC BACC30BL6)							
				(OPT HST79-6 (V56878))							
				(OPT HST79-6 (V92215))							
				(OPT HST79-6 (V5M902))							
168	NAS1399D5A2		.	RIVET						A, B, E, F	12
172	BACR15GF6D		.	RIVET						A, B, E, F	7
				(SIZE DETERMINED ON INST)							
176	BACB30NT3K7		.	BOLT						A, B, E, F	11
180	NAS1149D0316J		.	WASHER						A, B, E, F	11
184	BACR15GE3CW		.	RIVET						A, B, E, F	22
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 184A)							
-184A	AF5141-3C		.	RIVET						A, B, E, F	22
				(V53551)							
				(SPEC BACR15DR3AC)							
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 184)							
188	BACN10JP3ACD		.	NUTPLATE						A, B, E, F	2
192	MF51594-3-3BAC		.	NUTPLATE						A, B, E, F	2
				(V15653)							
				(SPEC BACN10YF33CD)							
196	MF51594-3-4BAC		.	NUTPLATE						A, B, E, F	7
				(V15653)							
				(SPEC BACN10YF34CD)							
200	114A7411-1		.	RIB ASSY-OUTBD END SS 517.77						A, E	1
				(OPT ITEM 200A)							
-200A	114A7411-5		.	RIB ASSY-OUTBD END SS 517.77						A, E	1
				(OPT ITEM 200)							
-204	114A7411-2		.	RIB ASSY-OUTBD END SS 517.77						B, F	1
				(OPT ITEM 204A)							
-204A	114A7411-6		.	RIB ASSY-OUTBD END SS 517.77						B, F	1
				(OPT ITEM 204)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
208	114A7412-1		.	RIB	ASSY-INBD	END	SS	384.97		A, E	1
-212	114A7412-2		.	RIB	ASSY-INBD	END	SS	384.97		B, F	1
216	BACR15BA3AD		.	RIVET						A, B, E, F	6
					(SIZE DETERMINED ON INST)			(USED ON ITEMS 200, 200A, 204, 204A)			
220	BACN10JP4ACD		.	NUTPLATE						A, B, E, F	3
					(USED ON ITEMS 200, 200A, 204, 204A)						
224	BACB28U3C030		.	BUSHING						A, B, E, F	2
228	114A7411-3		.	RIB						A, E	1
					(USED ON ITEM 200)			(OPT ITEM 228A)			
-228A	114A7411-7		.	RIB						A, E	1
					(USED ON ITEM 200A)			(OPT ITEM 228)			
-232	114A7411-4		.	RIB						B, F	1
					(USED ON ITEM 204)			(OPT ITEM 232A)			
-232A	114A7411-8		.	RIB						B, F	1
					(USED ON ITEM 204A)			(OPT ITEM 232)			
236	114A7412-3		.	RIB						A, E	1
					(USED ON ITEM 208)						
-240	114A7412-4		.	RIB						B, F	1
					(USED ON ITEM 212)						
244	BACS40R009C010F		.	SHIM						A, B, E, F	AR
248	BACS40R008B009F		.	SHIM						A, B, E, F	AR
252	BACS40R010C011F		.	SHIM						A, B, E, F	AR
256	114A9104-1		.	SEAL-SPONGE						A, B, E, F	2
260	BACB30NT3K3		.	BOLT						A, B, E, F	6
264	BACB30NT3K7		.	BOLT						A, B, E, F	2
268	NAS1149D0316J		.	WASHER						A, B, E, F	8
272	BACR15GE3CW		.	RIVET						A, B, E, F	16
					(SIZE DETERMINED ON INST)			(OPT ITEM 272A)			

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
1- -272A	AF5141-3C		.	R	I	V	E	T			A, B, E, F	16
			.	N	U	T	P	L	A	T	A	6
276	BACN10JP3ACD		.	N	U	T	P	L	A	T	A	1
280	MF51594-3-4BAC											
			.	N	U	T	P	L	A	T	A	1
284	MF51594-3-3BAC											
			.	R	E	T	A	I	N	E	R	1
288	114A9102-22		.	R	E	T	A	I	N	E	R	1
292	114A9102-23		.	R	E	T	A	I	N	E	R	1
-296	10-60754-1143											
296A	BA202179		.	S	E	A	L	-	B	U	L	1
-300	10-60754-1144											
-300A	BA202180		.	S	E	A	L	-	B	U	L	1
-304	10-60754-1145											
304A	BA202181		.	S	E	A	L	-	B	U	L	1
-308	10-60754-1146											
-308A	BA202182		.	S	E	A	L	-	B	U	L	1
312	BACR15FT5AD		.	R	I	V	E	T			A, B, E, F	4
316	114A5101-3		.	T	A	R	G	E	T	A	S	1
-320	114A5101-4		.	T	A	R	G	E	T	A	S	1
324	114A5101-5		.	T	A	R	G	E	T	A	S	1
-328	114A5101-6		.	T	A	R	G	E	T	A	S	1
-332	BACS12GX08H7											

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
332A	BACB30LH2K3		.	.						A, B, E, F	2
336	NAS1149DN816J		.	.						A, B, E, F	2
340	H52732-08CD		.	.						A, B, E, F	2
344	114A5108-2		.	.						A, E	1
-348	114A5108-1		.	.						B, F	1
352	114A5108-3		.	.						A, E	1
-356	114A5108-4		.	.						B, F	1
360	114A5107-4		.	.						A, B, E, F	1
364	114A5103-1		.	.						A, B, E, F	1
368	BACB30VF3K3		.							A, B, E, F	8
372	BACB30NT3K4		.							A, B, E, F	8
-376	NAS1149D0332J										
376A	NAS1149D0363J		.							A, B, E, F	16
380	BACW10P42S		.							A, B, E, F	8
384	H52732-3CD		.							A, B, E, F	16
388	114A5104-3		.							A, E	1
-392	114A5104-4		.							B, F	1
396	114A5104-1		.							A, E	1
-400	114A5104-2		.							B, F	1
404	114A5105-12		.							A, B, E, F	1
408	114A5105-11		.							A, B, E, F	1
412	114A5105-10		.							A, B, E, F	1
416	114A5105-9		.							A, B, E, F	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE	USAGE CODE	UNITS PER ASSY
1- 420	BACR15FT8D		. RIVET (SIZE DETERMINED ON INST) (OPT ITEM 420A)	A, B, E, F	4
-420A	BACR15FT8AD		. RIVET (SIZE DETERMINED ON INST) (OPT ITEM 420)	A, B, E, F	4
424	HST11AG6-5		. BOLT (V06725) (SPEC BACB30VU6K5) (OPT HST11AG6-5 (V73197)) (OPT HST11AG6-5 (V56878)) (OPT HST11AG6-5 (V0PTK6))	A, B, E, F	8
-428	HST11AG6-6		DELETED		
432	HST11AG6-4		. BOLT (V06725) (SPEC BACB30VU6K4) (OPT HST11AG6-4 (V73197)) (OPT HST11AG6-4 (V56878)) (OPT HST11AG6-4 (V0PTK6))	A, B, E, F	26
436	HST10AG6-3		. BOLT (V0PTK6) (SPEC BACB30VT6K3) (OPT HST10AG6-3 (V06725)) (OPT HST10AG6-3 (V56878)) (OPT HST10AG6-3 (V73197))	A, B, E, F	24
440	HST79CY6		. COLLAR (V73197) (SPEC BACC30BL6) (OPT HST79-6 (V56878)) (OPT HST79-6 (V92215)) (OPT HST79-6 (V5M902))	A, B, E, F	58
444	BACR15GF6D		. RIVET (SIZE DETERMINED ON INST)	A, B, E, F	30
448	BACB30NT3K7		. BOLT	A, B, E, F	44
452	NAS1149D0316J		. WASHER	A, B, E, F	44
456	BACR15GE3CW		. RIVET (SIZE DETERMINED ON INST) (OPT ITEM 456A)	A, B, E, F	88

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
1- -456A	AF5141-3C		.	R	I	V	E	T			A, B, E, F	88
			.	N	U	T	P	L	A	T	A	8
460	BACN10JP3ACD		.	N	U	T	P	L	A	T	A, B, E, F	8
464	MF51594-3-3BAC		.	N	U	T	P	L	A	T	A, B, E, F	27
468	MF51594-3-4BAC		.	N	U	T	P	L	A	T	A, B, E, F	7
472	MF51594-3-2BAC		.	N	U	T	P	L	A	T	A, B, E, F	2
476	114A7111-1		.	R	I	B	A	S	S	Y	A, E	1
-480	114A7111-2		.	R	I	B	A	S	S	Y	B, F	1
484	114A7112-1		.	R	I	B	A	S	S	Y	A, E	1
-488	114A7112-2		.	R	I	B	A	S	S	Y	B, F	1
492	114A7011-1		.	R	I	B	A	S	S	Y	A, E	1
-496	114A7011-2		.	R	I	B	A	S	S	Y	B, F	1
500	114A7012-1		.	R	I	B	A	S	S	Y	A, E	1
-504	114A7012-2		.	R	I	B	A	S	S	Y	B, F	1
508	ADW08V301NC		.	.	B	E	A	R	I	N	A, B, E, F	1
512	114A7011-3		.	.	R	I	B				A, E	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-516	114A7011-4		.	.	RIB (USED ON ITEM 496)					B, F	1
520	114A7012-3		.	.	RIB (USED ON ITEM 500)					A, E	1
-524	114A7012-4		.	.	RIB (USED ON ITEM 504)					B, F	1
528	BACS40R010C017F		.		SHIM					A, B, E, F	AR
532	BACS40R009B017F		.		SHIM					A, B, E, F	AR
536	BACS40R008B018F		.		SHIM					A, B, E, F	AR
540	BACS40R008B020F		.		SHIM					A, B, E, F	AR
544	BACS40R010C020F		.		SHIM					A, B, E, F	AR
548	BACS40R010C030F		.		SHIM					A, B, E, F	AR
552	HLT422AP8-13		.		BOLT (V97928) (SPEC BACB30YK8K13) (OPT HLT422AP8-13 (V06725)) (OPT HLT422AP8-13 (V60516))					A, B, E, F	2
556	HLT422AP10-16		.		BOLT (V97928) (SPEC BACB30YK10K16) (OPT HLT422AP10-16 (V06725)) (OPT HLT422AP10-16 (V60516)) (OPT ITEM 556A)					A, B, E, F	2
-556A	HLT422-10-16		.		BOLT (V73197) (OPT ITEM 556)					A, B, E, F	2
560	BACC30CP8S		.		COLLAR (OPT ITEM 561)					A, B, E, F	2
561	BACN10YT4CS		.		NUT (OPT ITEM 560)					A, B, E, F	2
564	BACC30CP10S		.		COLLAR (OPT ITEM 565)					A, B, E, F	2
565	BACN10YT5CS		.		NUT (OPT ITEM 564)					A, B, E, F	2
568	114A5107-2		.		SHIM-UPSTOP					A, B, E, F	2
572	114A7702-1		.		FITTING-UPSTOP					A, B, E, F	2
576	BACB30NT3K7		.		BOLT					A, B, E, F	2
580	NAS1149D0316J		.		WASHER					A, B, E, F	2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
584	BACR15GE3CW		.	R	I	V	E	T		A, B, E, F	4
-584A	AF5141-3C		.	R	I	V	E	T		A, B, E, F	4
-588	MF51594-3-3BAC										
588A	MF51594-3-2BAC		.	N	U	T	P	L	A	A, B, E, F	1
-592	BACN10JP3ACD										
592A	MF51594-3-4BAC		.	N	U	T	P	L	A	A, B, E, F	1
-596	114A5102-7										
596A	114A5102-17		.	R	E	T	A	I	N	A	A, E
-600	114A5102-8										
-600A	114A5102-18		.	R	E	T	A	I	N	A	B, F
604	BACR15FT8D		.	R	I	V	E	T		A, B, E, F	3
-604A	BACR15FT8AD		.	R	I	V	E	T		A, B, E, F	3
608	HST11AG6-5		.	B	O	L	T			A, B, E, F	6
-612	HST11AG6-6										
616	HST11AG6-4		.	B	O	L	T			A, B, E, F	20

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
1- 620	HST10AG6-3		.	BOLT							A, B, E, F	18
				(V0PTK6)								
				(SPEC BACB30VT6K3)								
				(OPT HST10AG6-3 (V06725))								
				(OPT HST10AG6-3 (V56878))								
				(OPT HST10AG6-3 (V73197))								
624	HST79CY6		.	COLLAR							A, B, E, F	44
				(V73197)								
				(SPEC BACC30BL6)								
				(OPT HST79-6 (V56878))								
				(OPT HST79-6 (V92215))								
				(OPT HST79-6 (V5M902))								
628	BACR15GF6D		.	RIVET							A, B, E, F	20
				(SIZE DETERMINED ON INST)								
632	BACB30NT3K7		.	BOLT							A, B, E, F	35
636	NAS1149D0316J		.	WASHER							A, B, E, F	35
640	BACR15GE3CW		.	RIVET							A, B, E, F	70
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 640A)								
-640A	AF5141-3C		.	RIVET							A, B, E, F	70
				(V53551)								
				(SPEC BACR15DR3AC)								
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 640)								
644	BACN10JP3ACD		.	NUTPLATE							A, B, E, F	6
648	MF51594-3-2BAC		.	NUTPLATE							A, B, E, F	12
				(V15653)								
				(SPEC BACN10YF32CD)								
652	MF51594-3-3BAC		.	NUTPLATE							A, B, E, F	17
				(V15653)								
				(SPEC BACN10YF33CD)								
656	114A7311-1		.	RIB ASSY-UPSTOP OUTBD SS 457.27							A, E	1
				(FOR DETAILS SEE FIG. 3)								
-660	114A7311-2		.	RIB ASSY-UPSTOP OUTBD SS 457.27							B, F	1
				(FOR DETAILS SEE FIG. 3)								
664	114A7312-1		.	RIB ASSY-UPSTOP INBD SS 446.67							A, E	1
				(FOR DETAILS SEE FIG. 3)								
-668	114A7312-2		.	RIB ASSY-UPSTOP INBD SS 446.67							B, F	1
				(FOR DETAILS SEE FIG. 3)								
672	114A7211-1		.	RIB ASSY-ACTR SS 451.97							A, E	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-676	114A7211-2		.	RIB	ASSY-AC	TR	SS	451.97		B, F	1
680	BACB28AP10P032		.	.	BUSHING					A, B, E, F	1
684	BACB28AT14B032C		.	.	BUSHING					A, B, E, F	1
688	114A7211-3		.	.	RIB					A, E	1
-692	114A7211-4		.	.	RIB					B, F	1
696	BACS40R010C017F		.	SHIM						A, B, E, F	AR
700	BACS40R009B017F		.	SHIM						A, B, E, F	AR
704	BACS40R008B020F		.	SHIM						A, B, E, F	AR
708	BACS40R008B018F		.	SHIM						A, B, E, F	AR
712	BACS40R010C020F		.	SHIM						A, B, E, F	AR
716	BACB30NT3K7		.	BOLT						A, B, E, F	52
720	BACB30NT3K3		.	BOLT						A, B, E, F	59
724	NAS1149D0316J		.	WASHER						A, B, E, F	111
728	BACR15GE3CW		.	RIVET						A, B, E, F	222
					(SIZE DETERMINED ON INST)						
					(OPT ITEM 728A)						
-728A	AF5141-3C		.	RIVET						A, B, E, F	222
					(V53551)						
					(SPEC BACR15DR3AC)						
					(SIZE DETERMINED ON INST)						
					(OPT ITEM 728)						
732	MF51594-3-3BAC		.	NUTPLATE						A, B, E, F	52
					(V15653)						
					(SPEC BACN10YF33CD)						
736	BACN10JP3ACD		.	NUTPLATE						A, B, E, F	59
-740	114A9102-1			DELETED							
740A	114A9102-30		.	RETAINER-SEAL						A, B, E, F	1
744	114A9102-2		.	RETAINER-SEAL						A, B, E, F	2
748	114A9102-3		.	RETAINER-SEAL						A, B, E, F	2
752	114A9102-4		.	RETAINER-SEAL						A, B, E, F	2
756	114A9102-5		.	RETAINER-SEAL						A, B, E, F	1
-760	114A9101-1			DELETED							
760A	114A9101-45		.	SEAL-BULB						A, E	1
					(MAKE FROM EXTR RUBBER						
					10-60754-482 X 24.36)						

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-764	114A9101-2										
-764A	114A9101-46									B, F	1
768	114A9101-3									A, E	1
-772	114A9101-4									B, F	1
776	114A9101-5									A, E	1
-780	114A9101-6									B, F	1
784	114A9101-7									A, B, E, F	2
788	114A9101-9									A, E	1
-792	114A9101-10									B, F	1
796	114A9101-11									A, E	1
-800	114A9101-12									B, F	1
804	114A9101-13									A, E	1
-808	114A9101-14									B, F	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY		
			1	2	3	4	5	6	7				
1- 812	HST11AG6-3		.	B	O	L	T				A, B, E, F	17	
				(V06725)	(SPEC BACB30VU6K3)	(OPT HST11AG6-3 (V73197))	(OPT HST11AG6-3 (V56878))	(OPT HST11AG6-3 (V0PTK6))					
816	HST79CY6		.	C	O	L	L	A	R		A, B, E, F	17	
				(V73197)	(SPEC BACC30BL6)	(OPT HST79-6 (V56878))	(OPT HST79-6 (V92215))	(OPT HST79-6 (V5M902))					
820	BACR15GF6D		.	R	I	V	E	T			A, B, E, F	263	
				(SIZE DETERMINED ON INST)									
824	BACR15BB8AD		.	R	I	V	E	T			A, B, E, F	5	
				(SIZE DETERMINED ON INST)									
828	114A6010-1		.	W	E	D	G	E	A	S	A	A, E	1
-832	114A6010-2		.	W	E	D	G	E	A	S	A	B, F	1
836	114A8211-1		.	B	E	A	M	-	N	O	S	A, E	1
-840	114A8211-2		.	B	E	A	M	-	N	O	S	B, F	1
842	NAS1399D4A2		.	R	I	V	E	T			E, F	4	
843	114A5109-1		.	V	O	R	T	I	L	O	N	E	1
				(OPT ITEM 843A)									
-843A	114A5109-7		.	V	O	R	T	I	L	O	N	E	1
				(OPT ITEM 843)									
-844	114A5109-2		.	V	O	R	T	I	L	O	N	F	1
				(OPT ITEM 844A)									
-844A	114A5109-8		.	V	O	R	T	I	L	O	N	F	1
				(OPT ITEM 844)									
845	BACR15BB3AD		.	R	I	V	E	T			A, B, E, F	4	
				(SIZE DETERMINED ON INST)									
850	MS27253-1		.	P	L	A	T	E			A, B, E, F	1	

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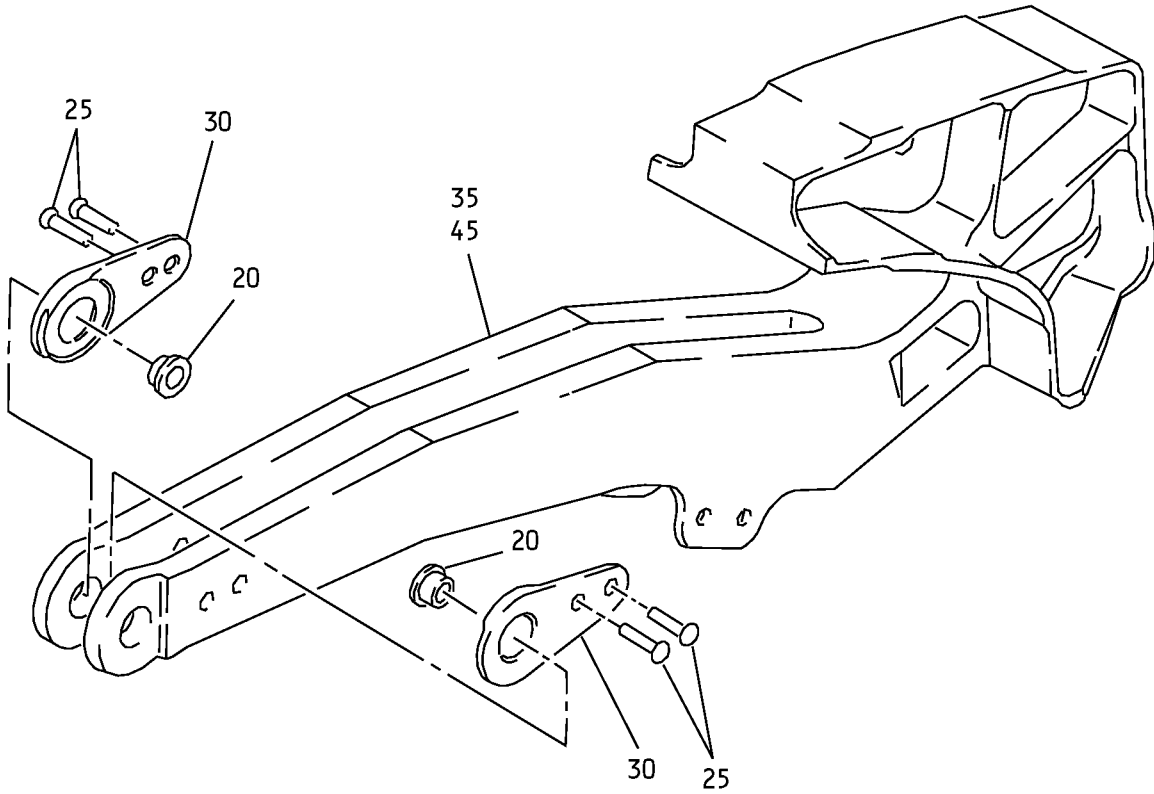
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Outboard/Inboard Auxiliary Arm Rib Assembly
IPL Figure 2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
2-												
-1A	114A7111-1									RIB ASSY-OUTBD AUX ARM SS 480.97	A, C, E, G, J	RF
-5	114A7111-2									RIB ASSY-OUTBD AUX ARM SS 480.97	B, D, F, H, K	RF
-10	114A7112-1									RIB ASSY-INBD AUX ARM SS 422.97	A, C, E, G, J	RF
-15	114A7112-2									RIB ASSY-INBD AUX ARM SS 422.97	B, D, F, H, K	RF
20	BACB28AP06P022									. BUSHING		2
25	BACR15BB5AD									. RIVET (SIZE DETERMINED ON INST)		4
30	114A7101-1									. GUARD-SIDE		2
35	114A7111-3									. RIB (USED ON ITEM 1A)	A, C, E, G, J	1
-40	114A7111-4									. RIB (USED ON ITEM 5)	B, D, F, H, K	1
45	114A7112-3									. RIB (USED ON ITEM 10)	A, C, E, G, J	1
-50	114A7112-4									. RIB (USED ON ITEM 15)	B, D, F, H, K	1

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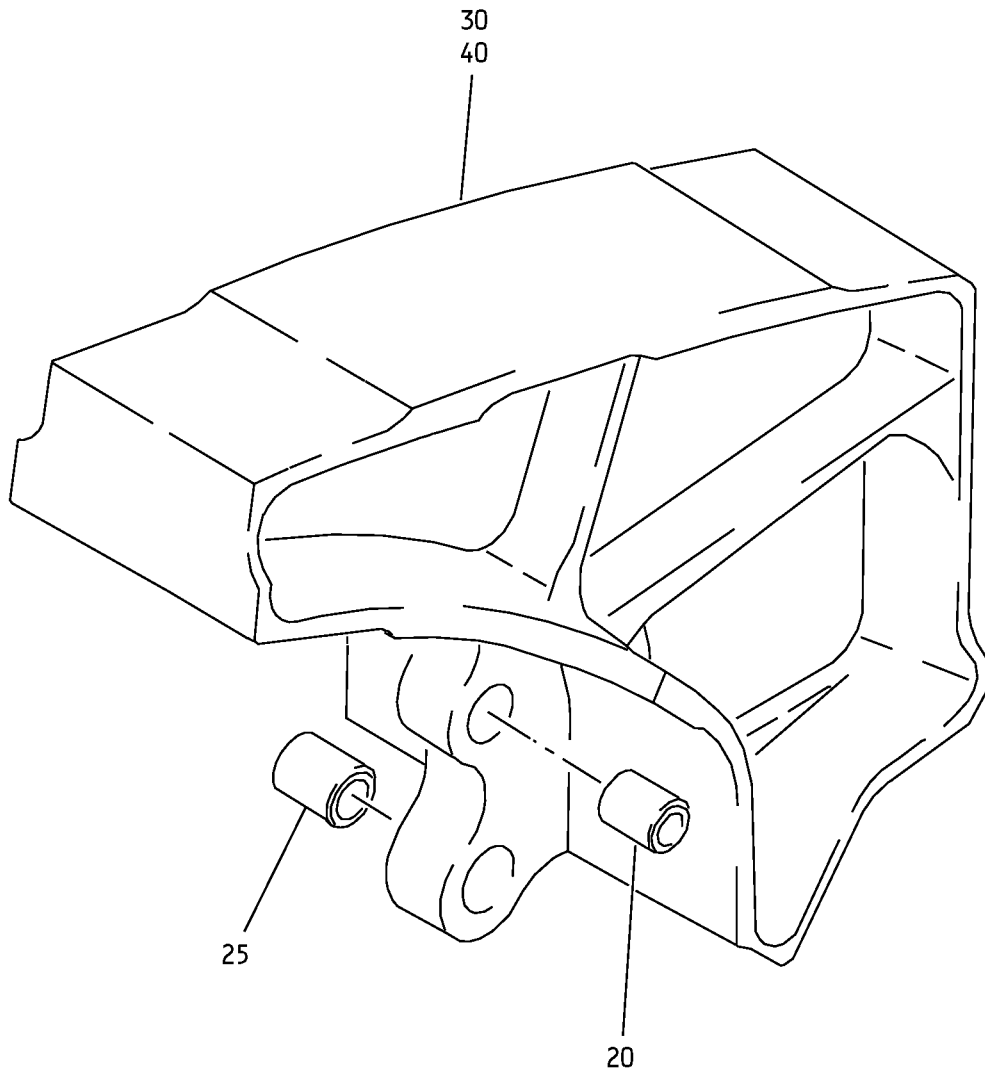
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Upstop Outboard/Inboard Rib Assembly
IPL Figure 3

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
3-												
-1A	114A7311-1									RIB ASSY-UPSTOP OUTBD SS 457.27	A, C, E, J	RF
-1B	114A7311-201									RIB ASSY-UPSTOP OUTBD SS 457.27	G	RF
-5	114A7311-2									RIB ASSY-UPSTOP OUTBD SS 457.27	B, D, F, K	RF
-5A	114A7311-202									RIB ASSY-UPSTOP OUTBD SS 457.27	H	RF
-10	114A7312-1									RIB ASSY-UPSTOP INBD SS 446.67	A, C, E, J	RF
-10A	114A7312-201									RIB ASSY-UPSTOP INBD SS 446.67	G	RF
-15	114A7312-2									RIB ASSY-UPSTOP INBD SS 446.67	B, D, F, K	RF
-15A	114A7312-202									RIB ASSY-UPSTOP INBD SS 446.67	H	RF
20	BAC~ B28AW04B044C									. BUSHING		1
25	BAC~ B28AW05B044C									. BUSHING		1
30	114A7311-3									. RIB (USED ON ITEM 1A)	A, C, E, J	1
-30A	114A7311-203									. RIB (USED ON ITEM 1B)	G	1
-35	114A7311-4									. RIB (USED ON ITEM 5)	B, D, F, K	1
-35A	114A7311-204									. RIB (USED ON ITEM 5A)	H	1
40	114A7312-3									. RIB (USED ON ITEM 10)	A, C, E, J	1
-40A	114A7312-203									. RIB (USED ON ITEM 10A)	G	1
-45	114A7312-4									. RIB (USED ON ITEM 15)	B, D, F, K	1
-45A	114A7312-204									. RIB (USED ON ITEM 15A)	H	1

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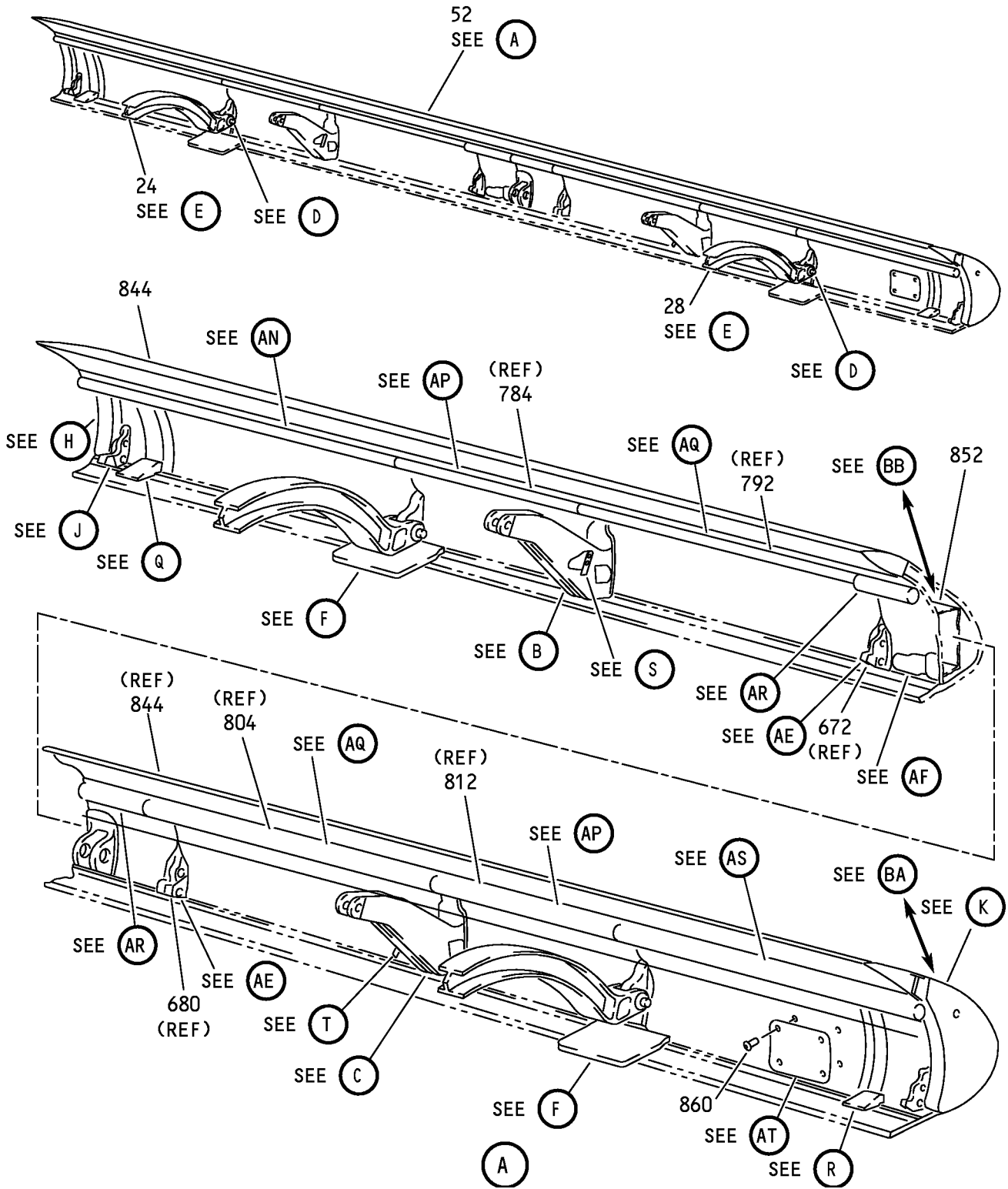
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 1 of 29)

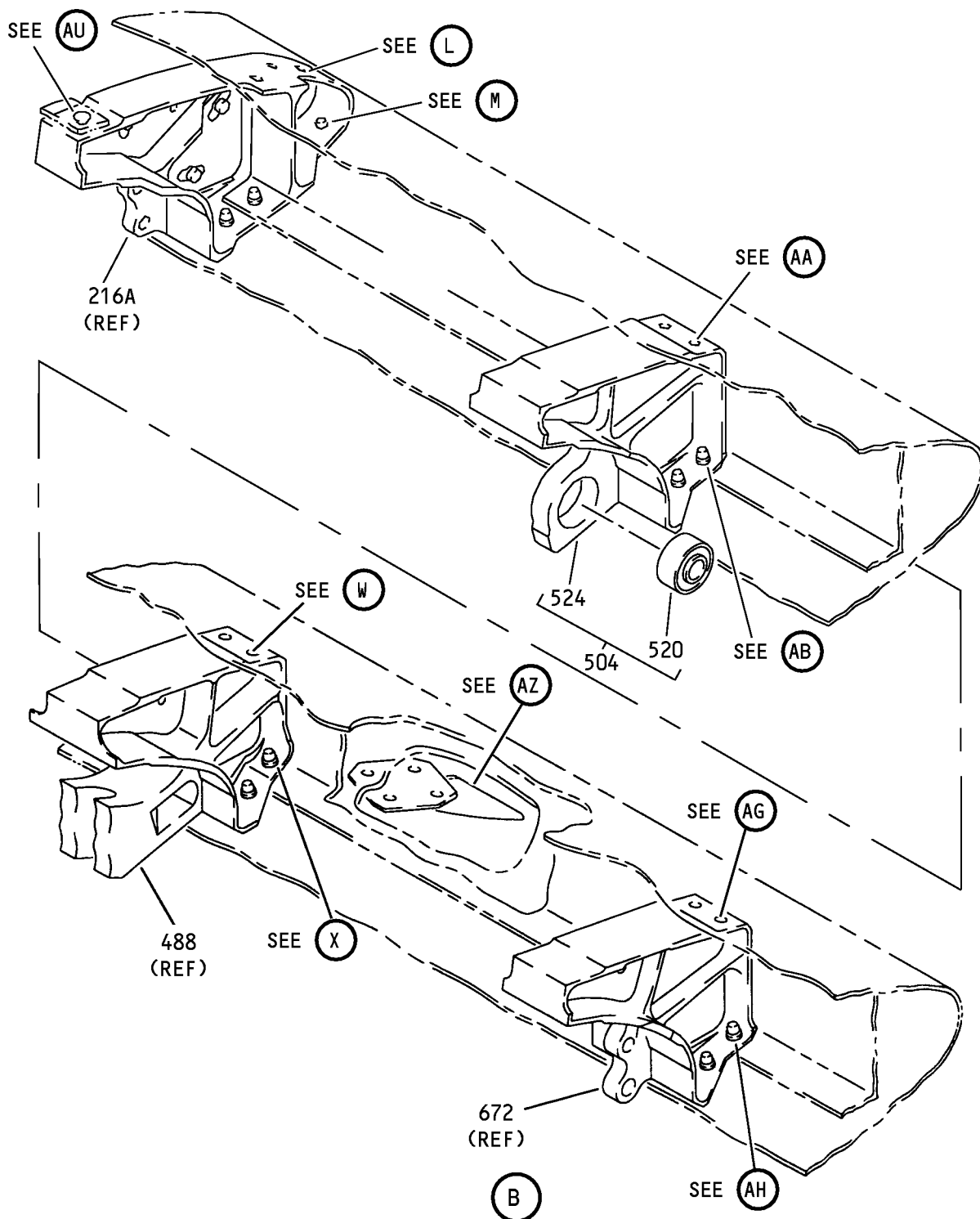
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 2 of 29)

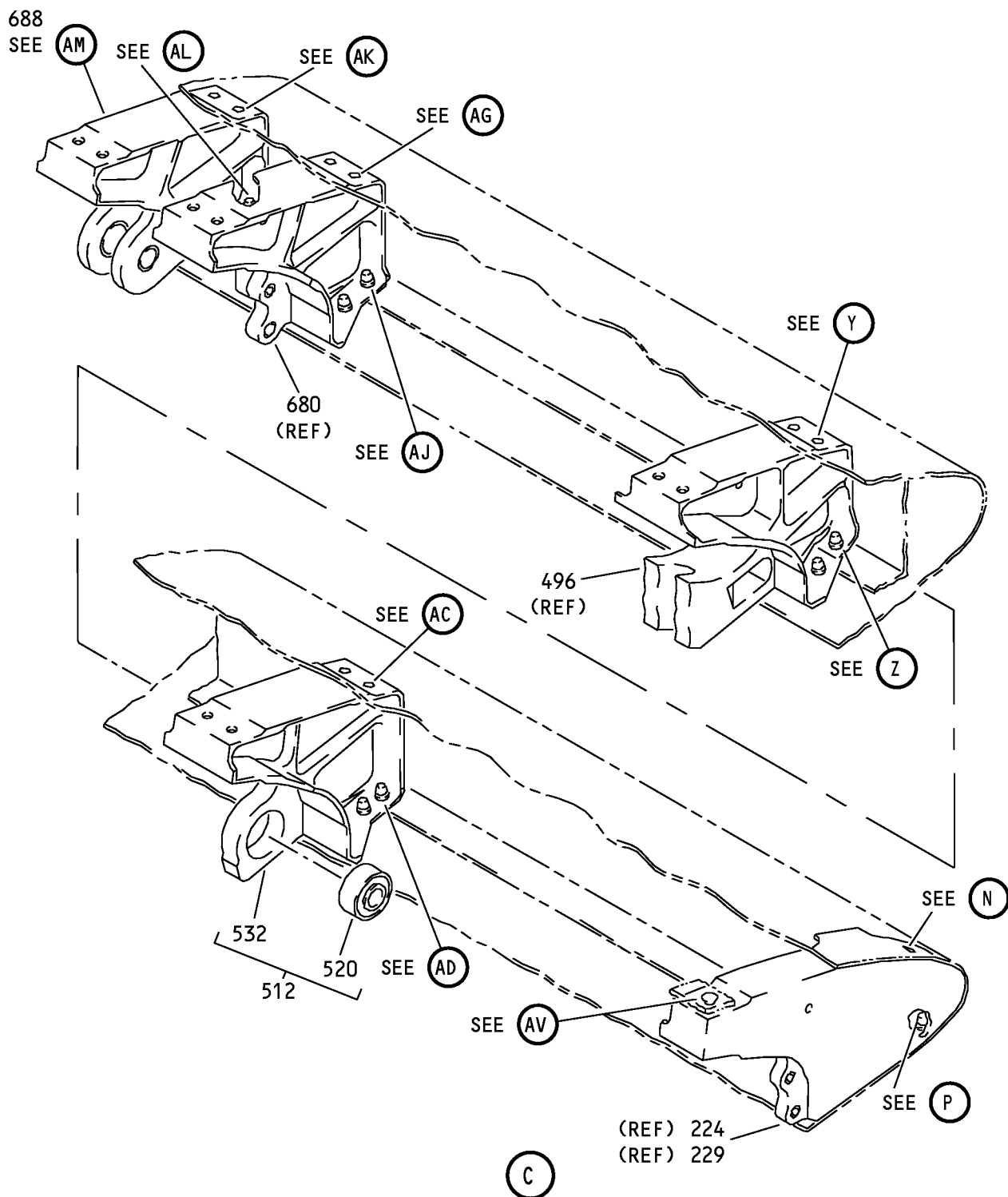
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 3 of 29)

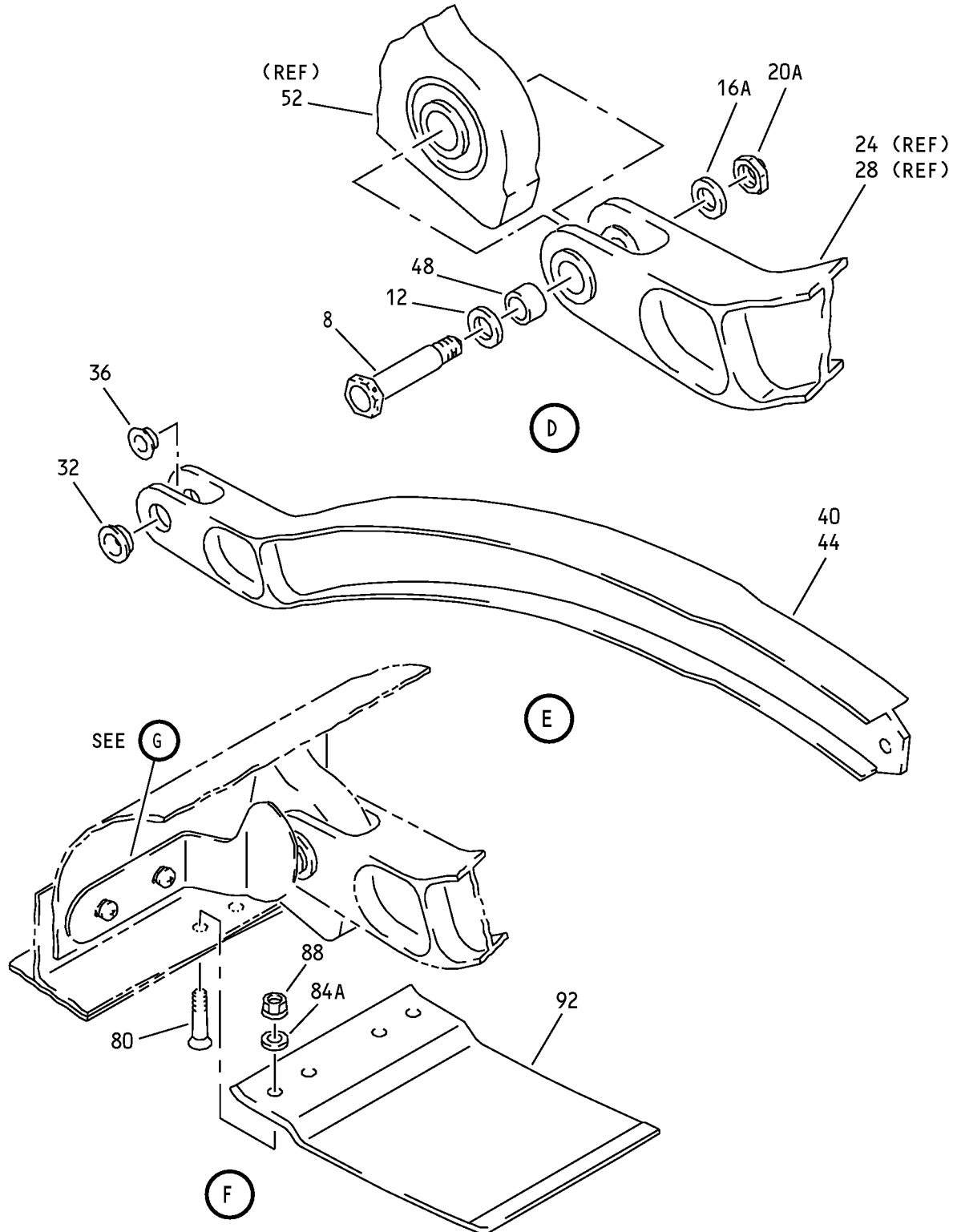
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 4 of 29)

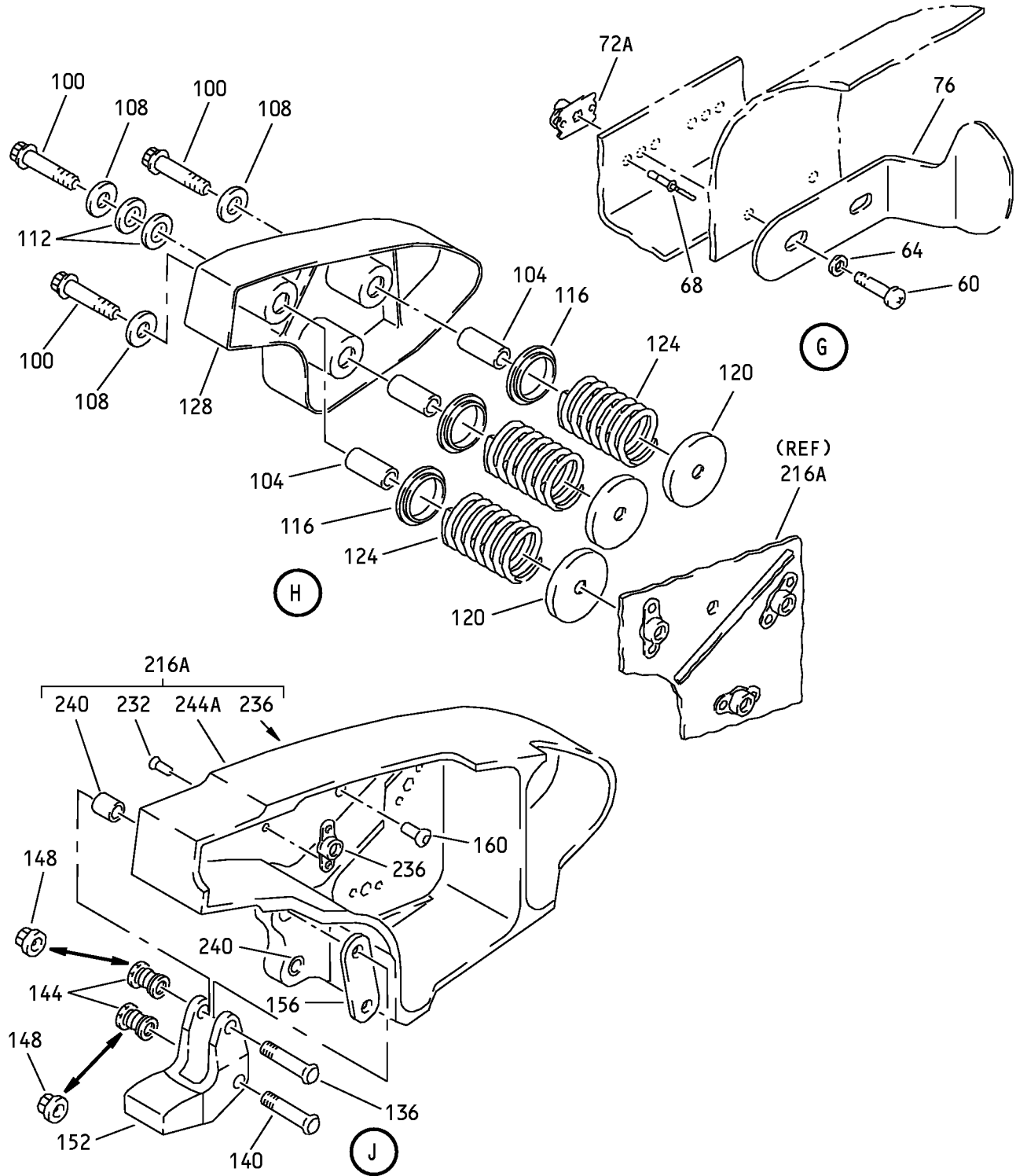
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 5 of 29)

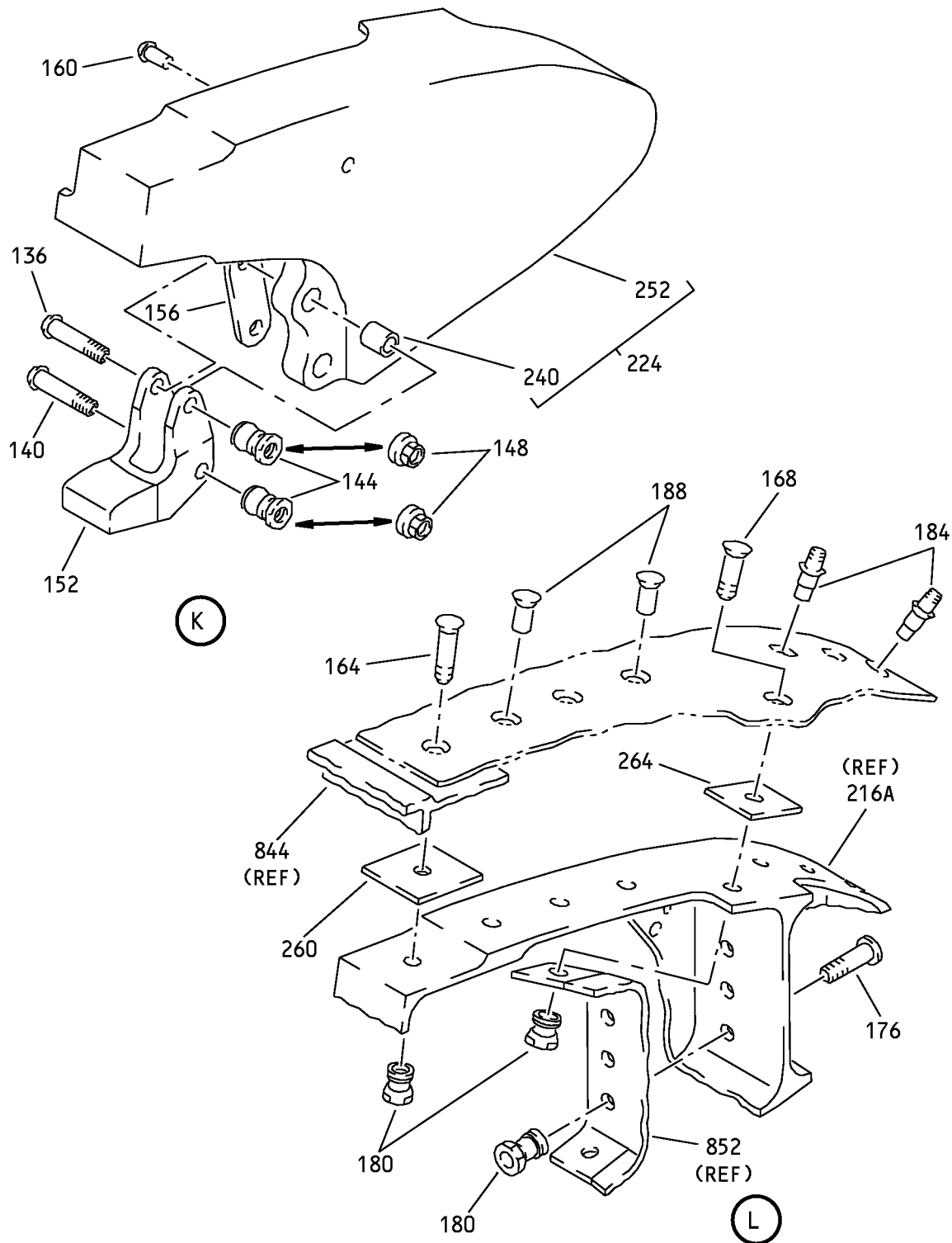
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 6 of 29)

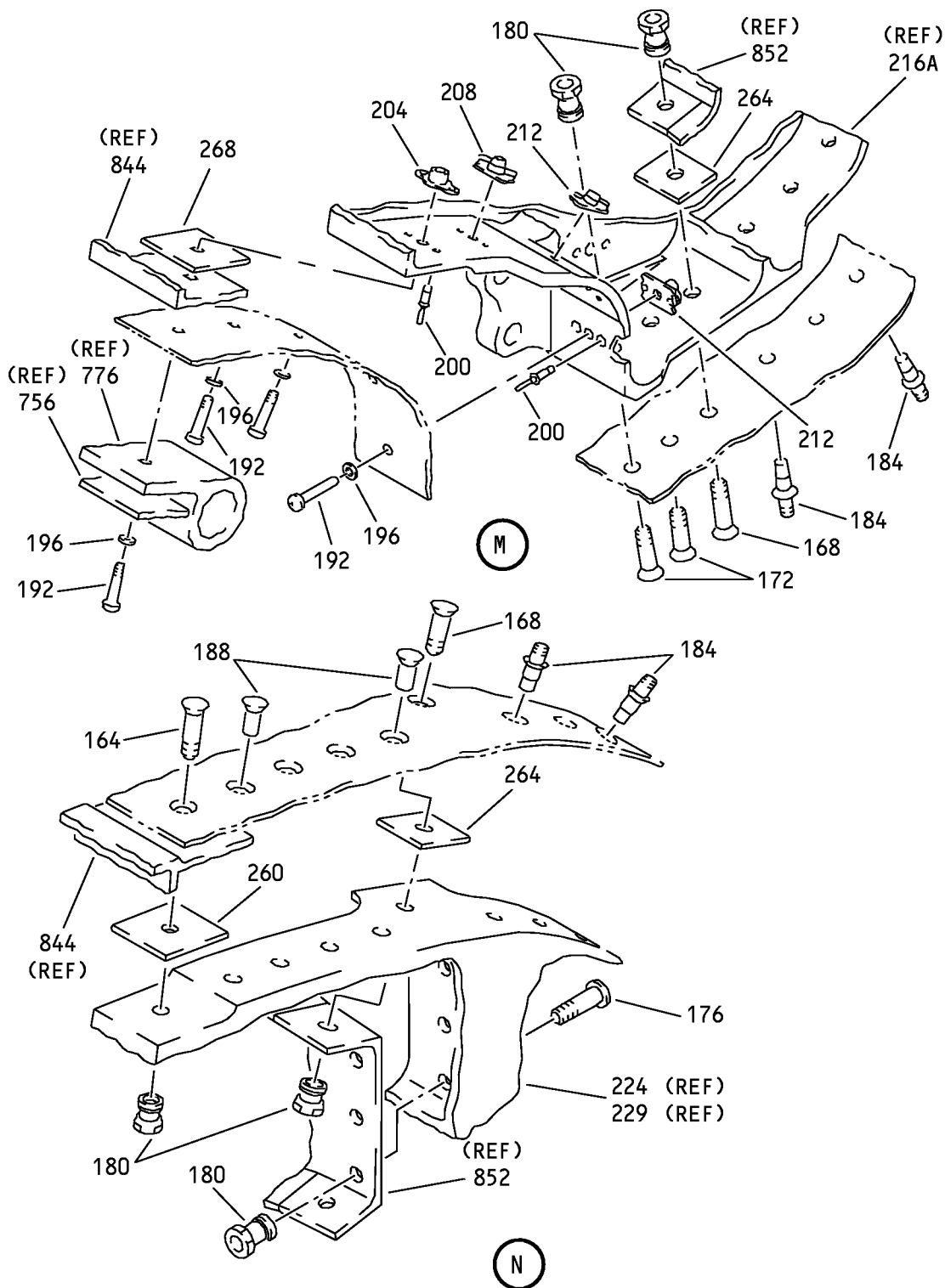
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 7 of 29)

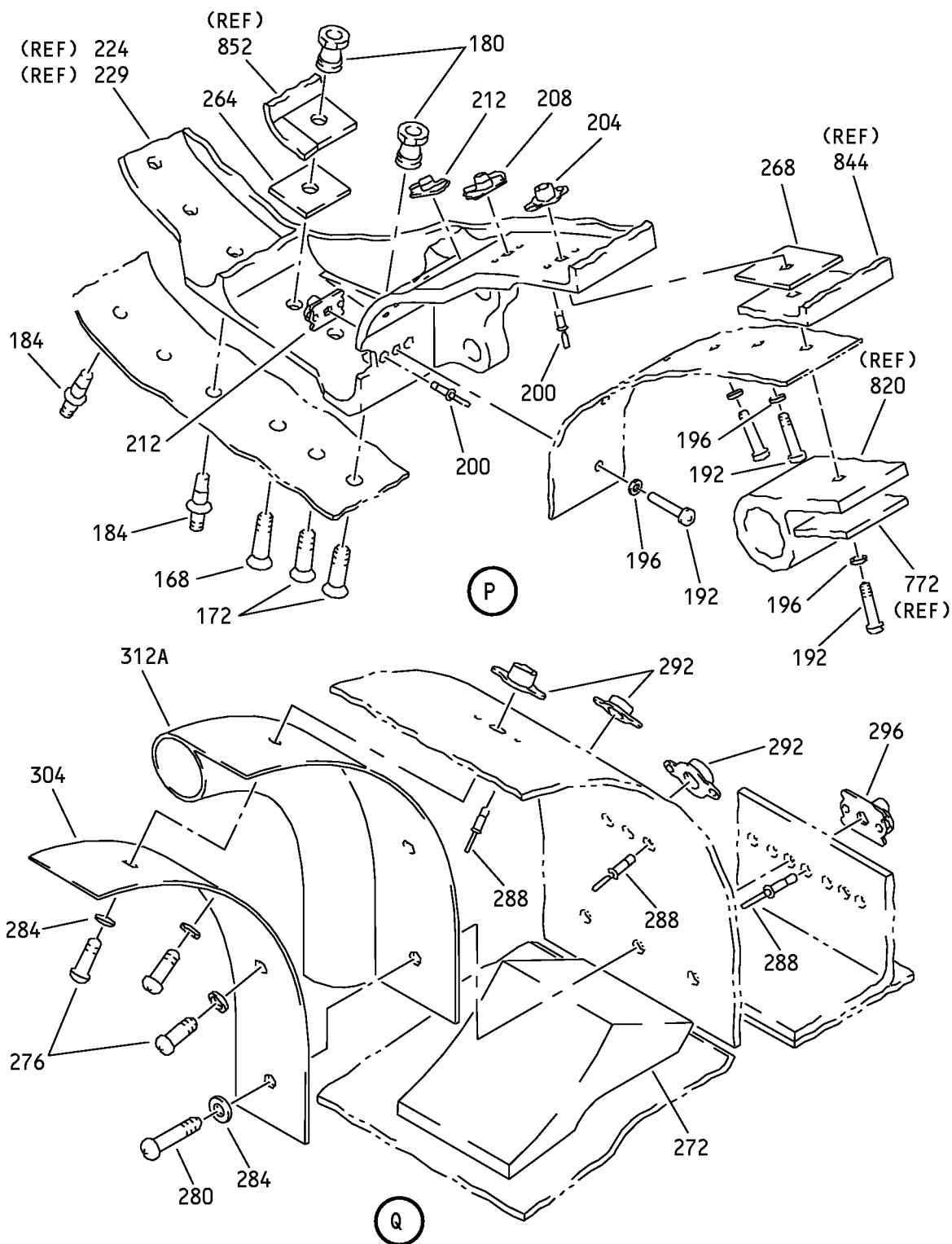
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 8 of 29)

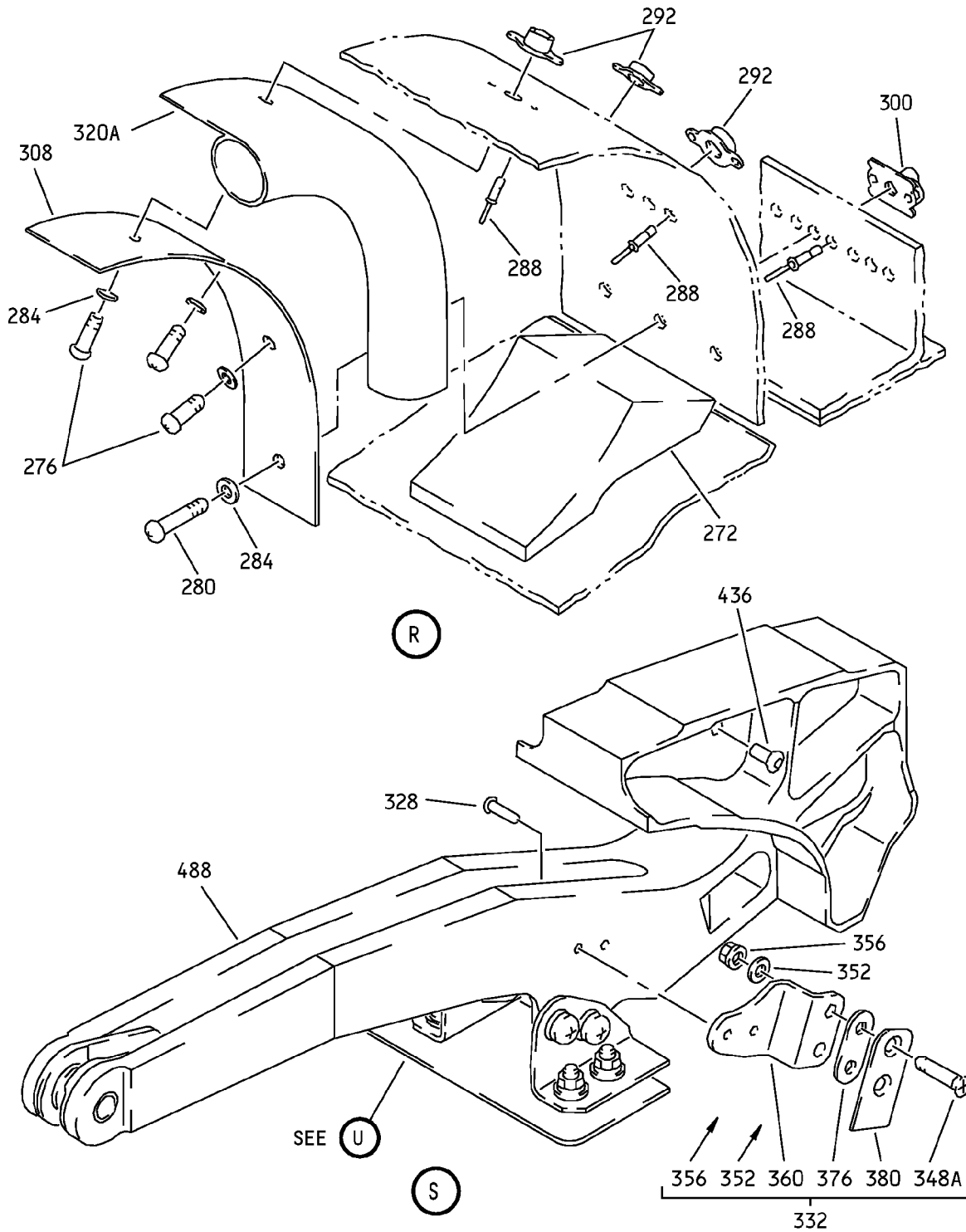
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 9 of 29)

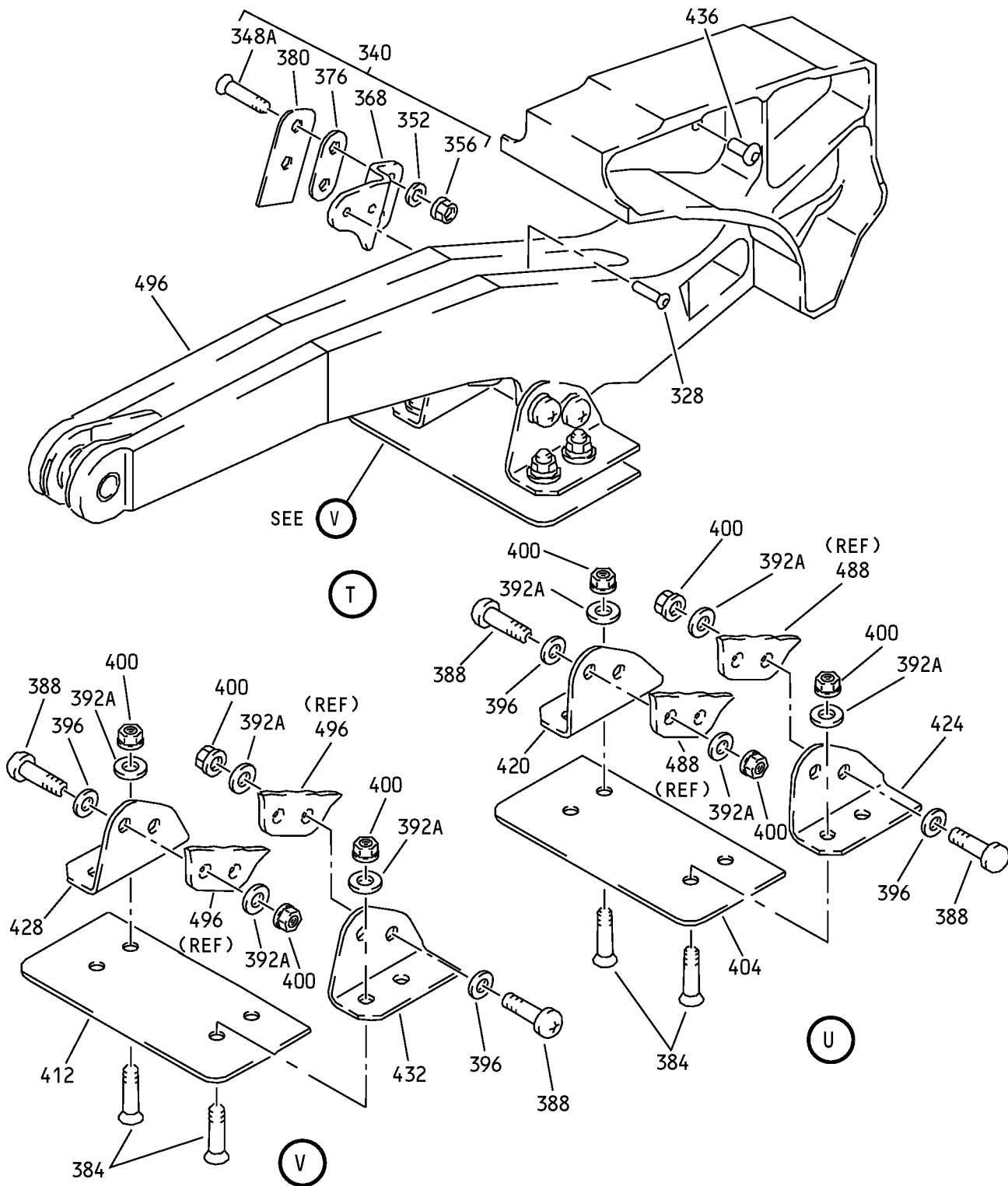
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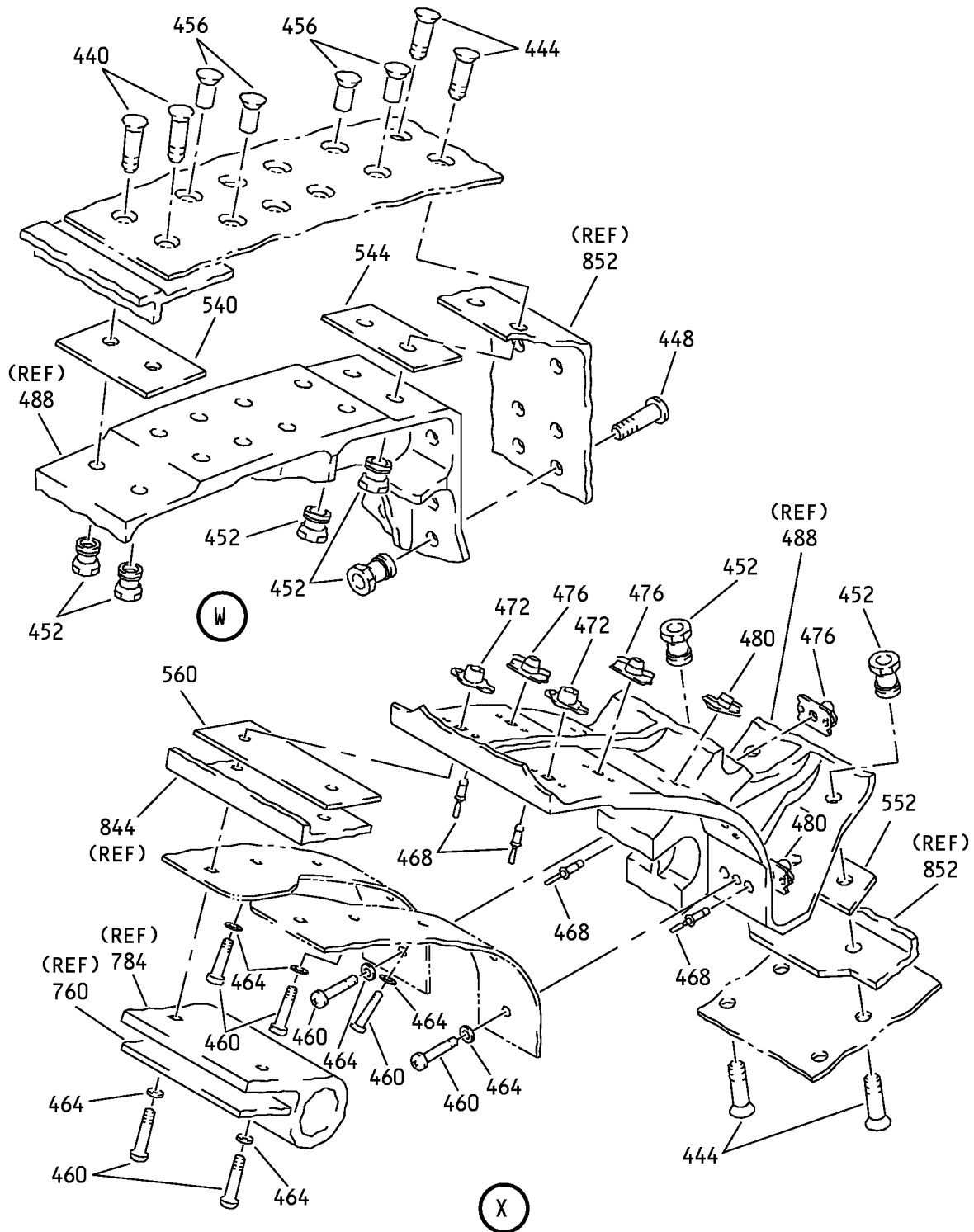
Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 10 of 29)

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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 11 of 29)

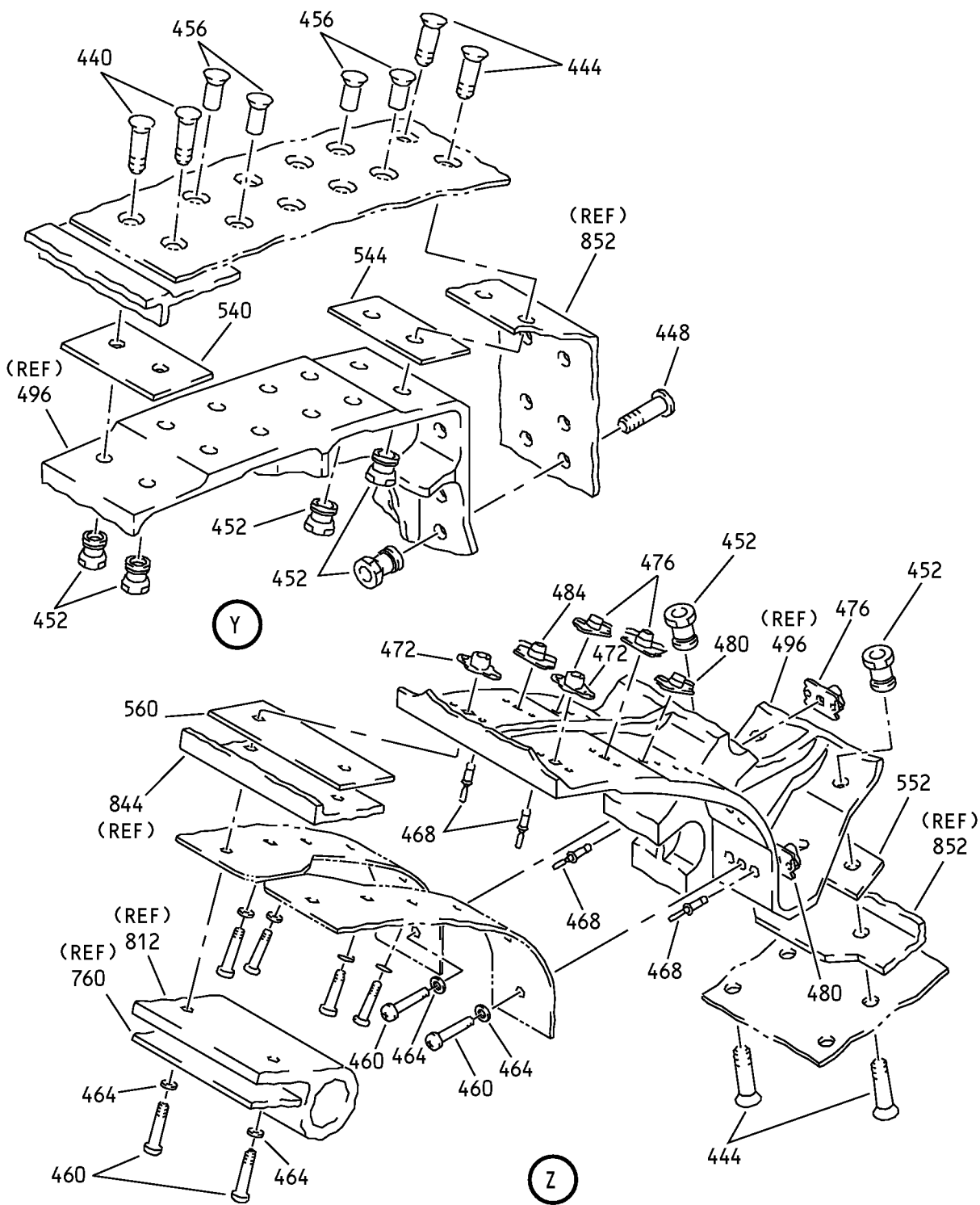
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 12 of 29)

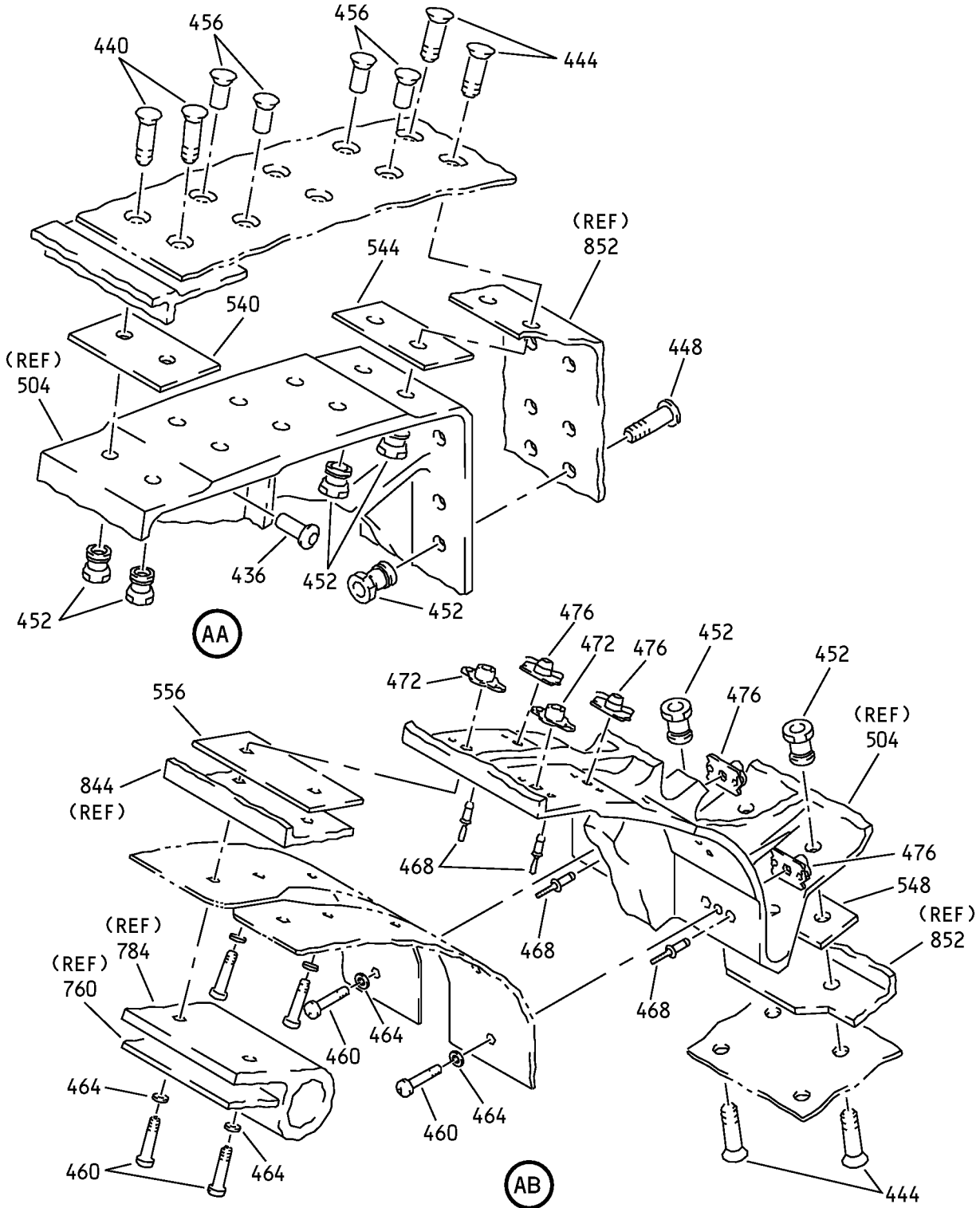
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 13 of 29)

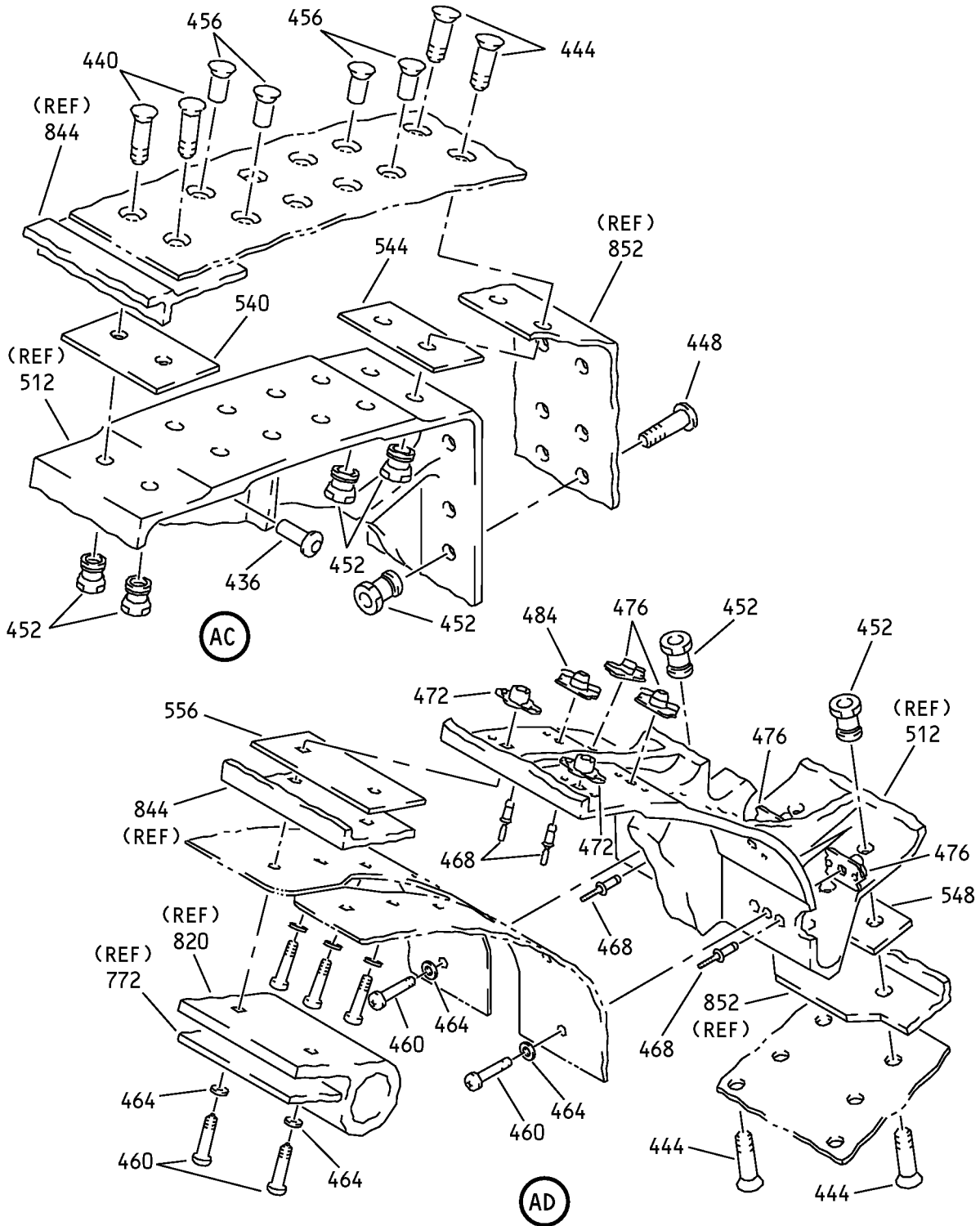
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 14 of 29)

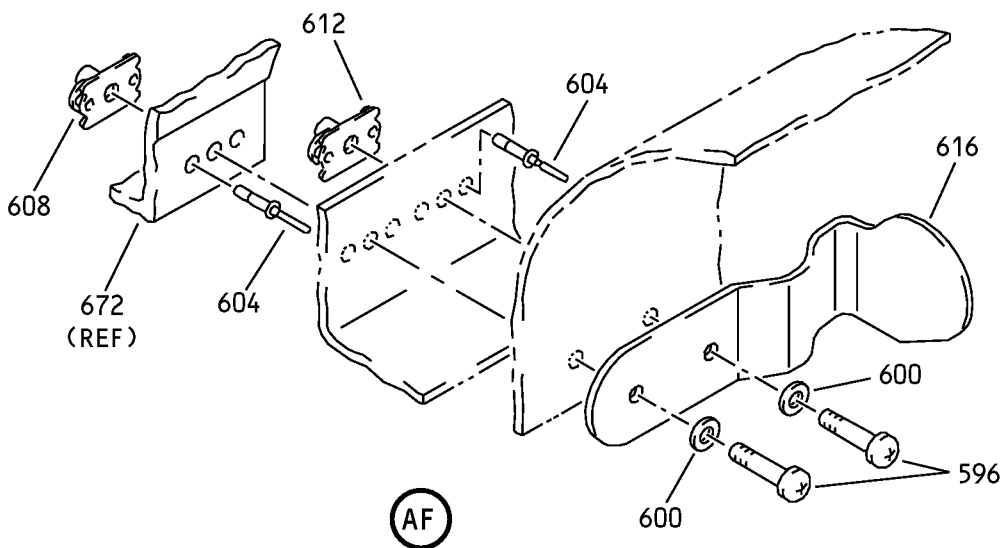
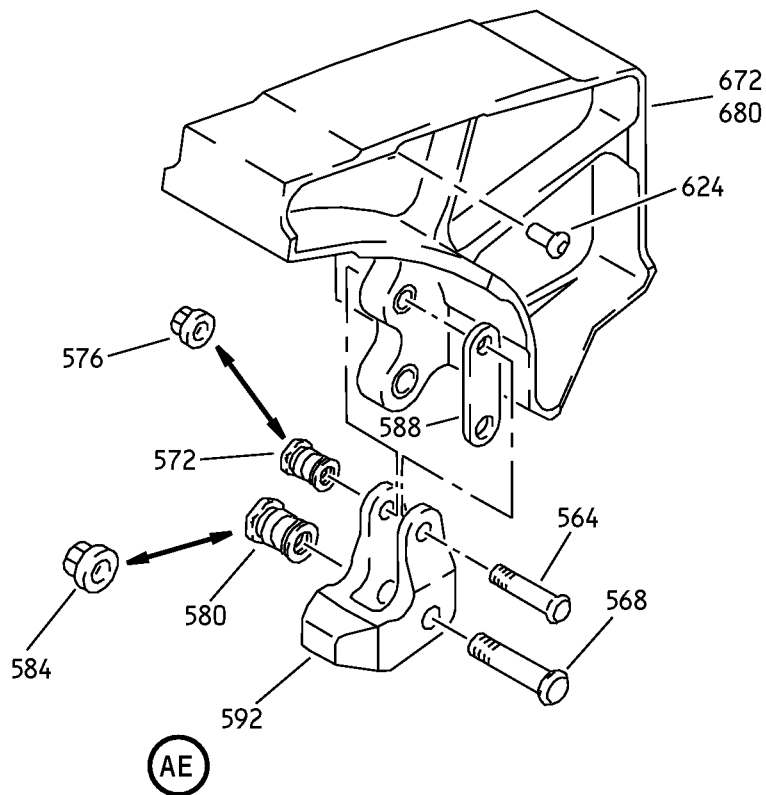
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 15 of 29)

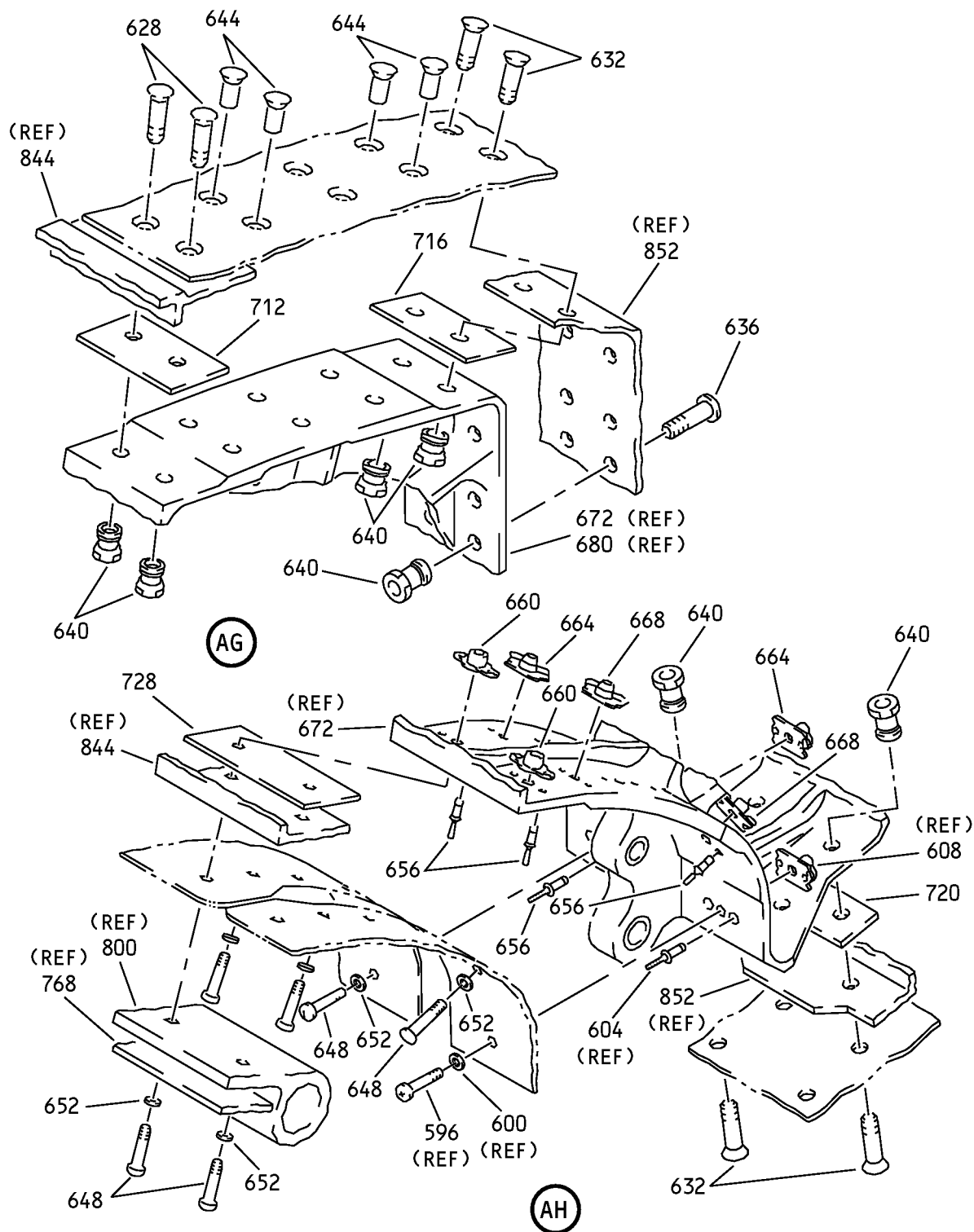
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 16 of 29)

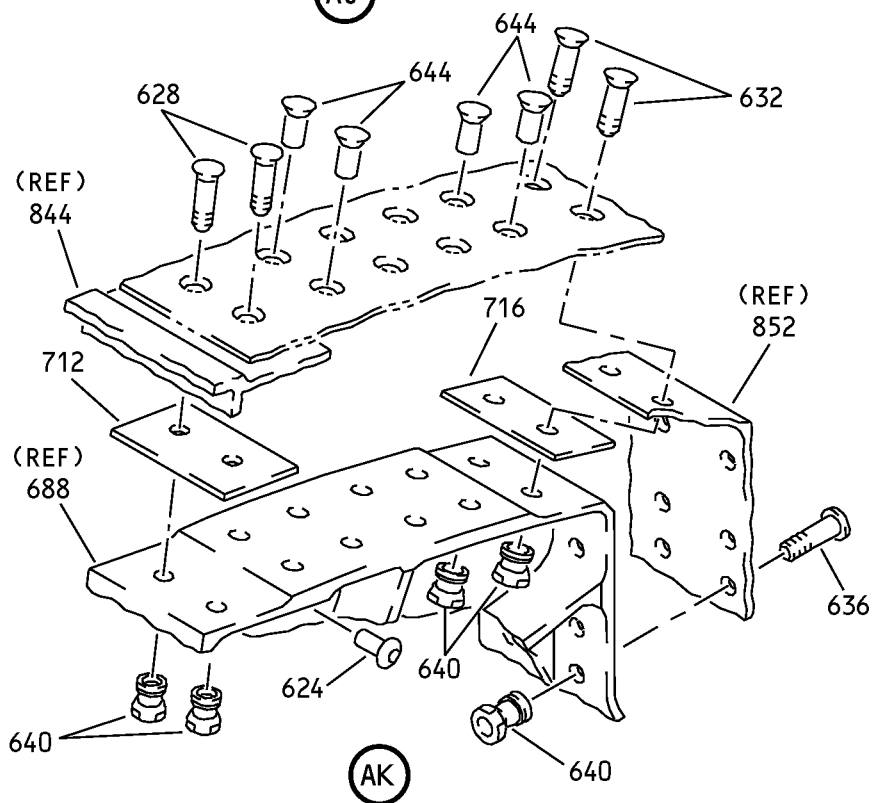
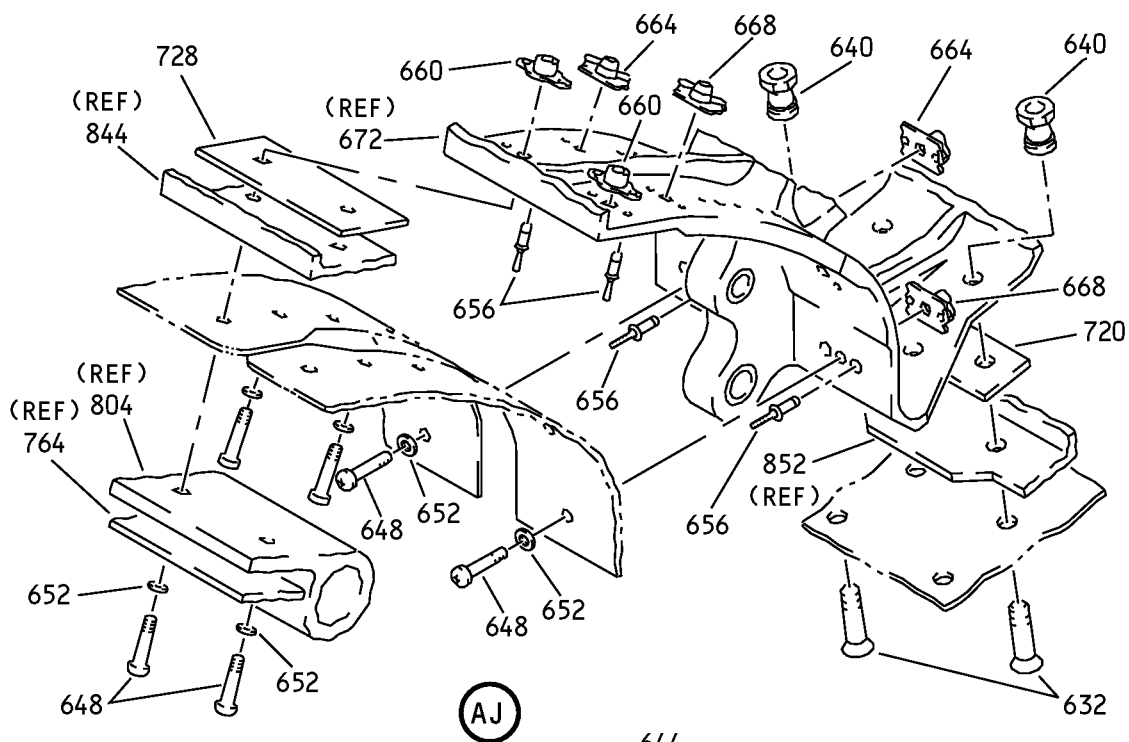
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 17 of 29)

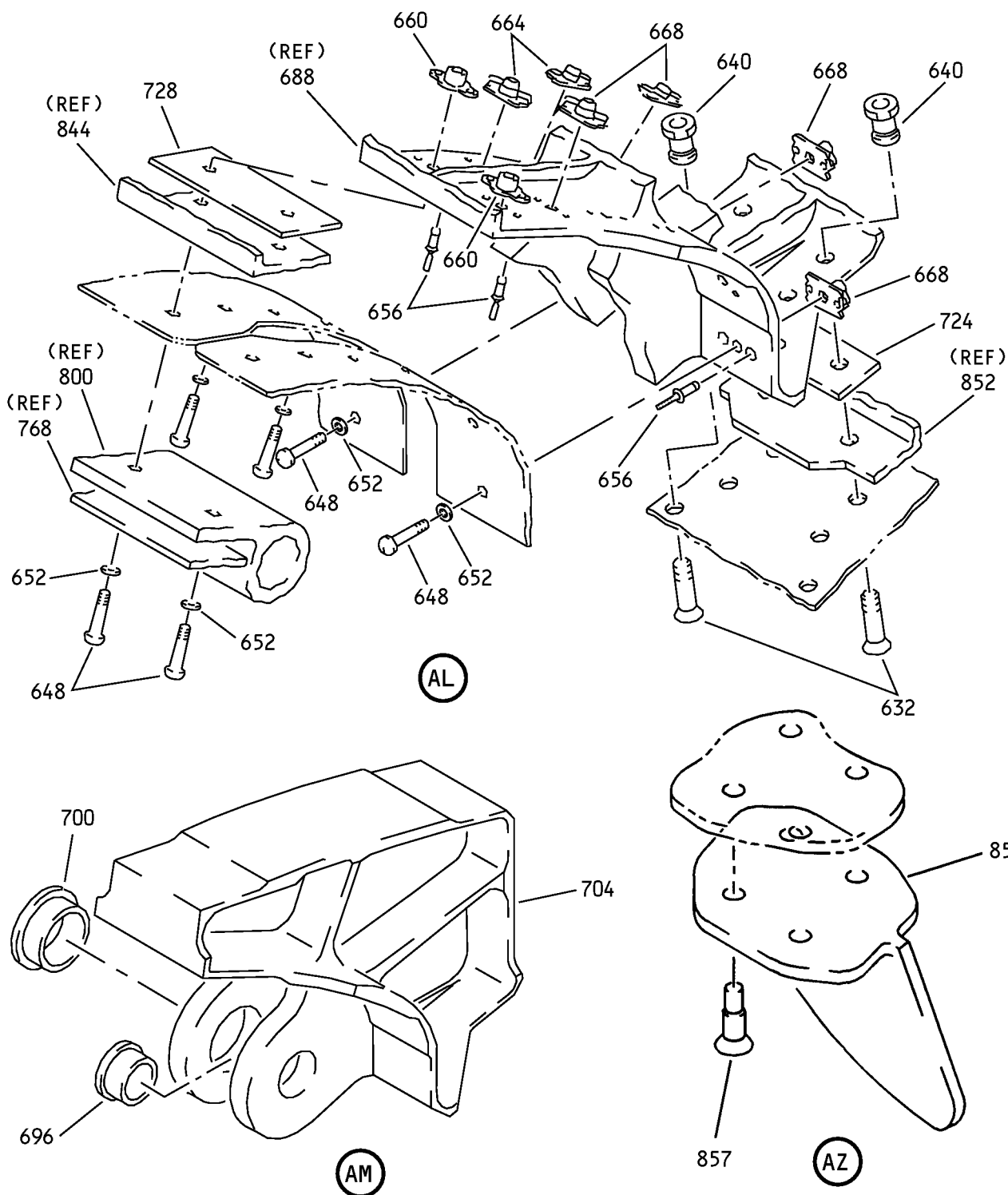
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 18 of 29)

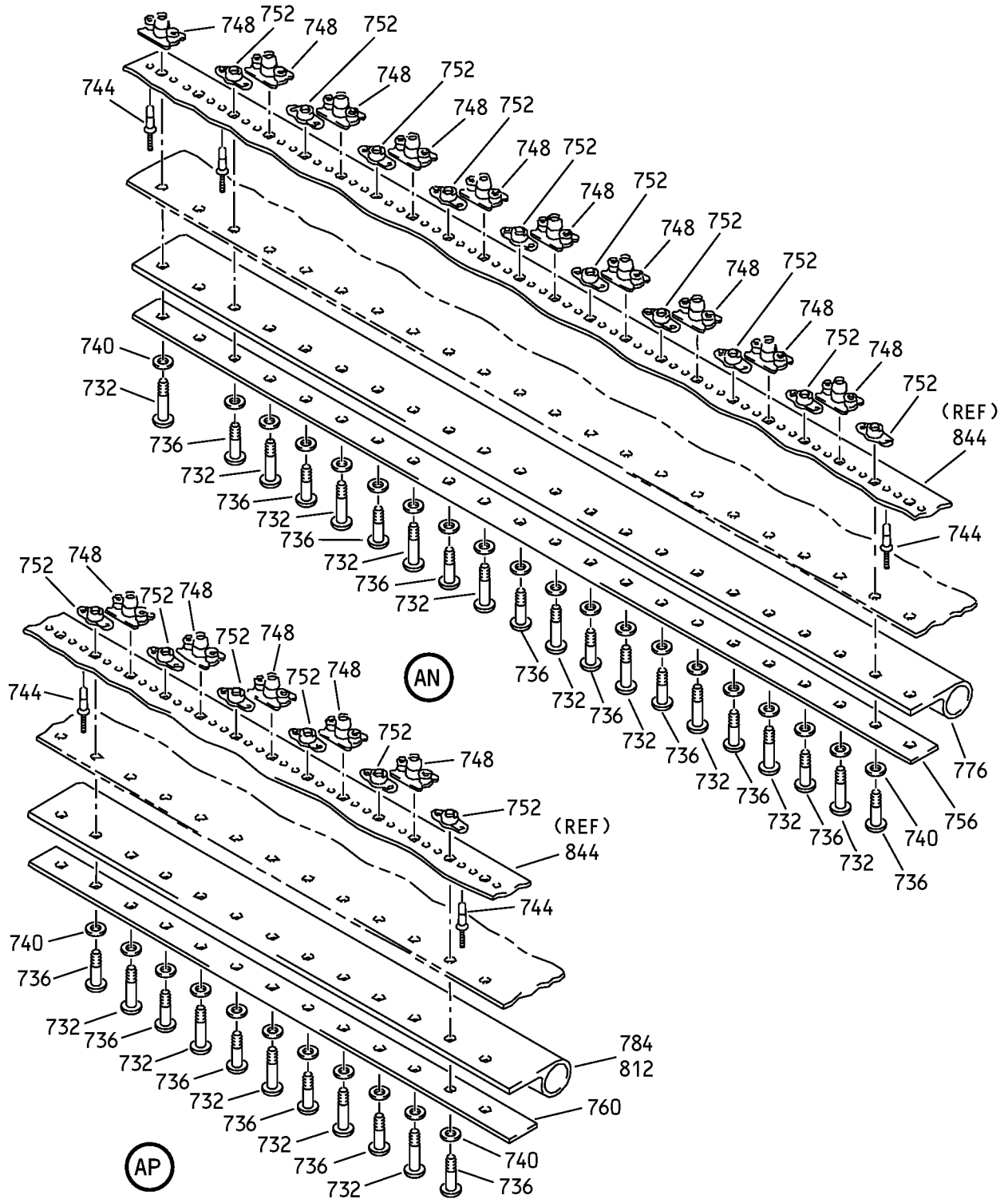
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 19 of 29)

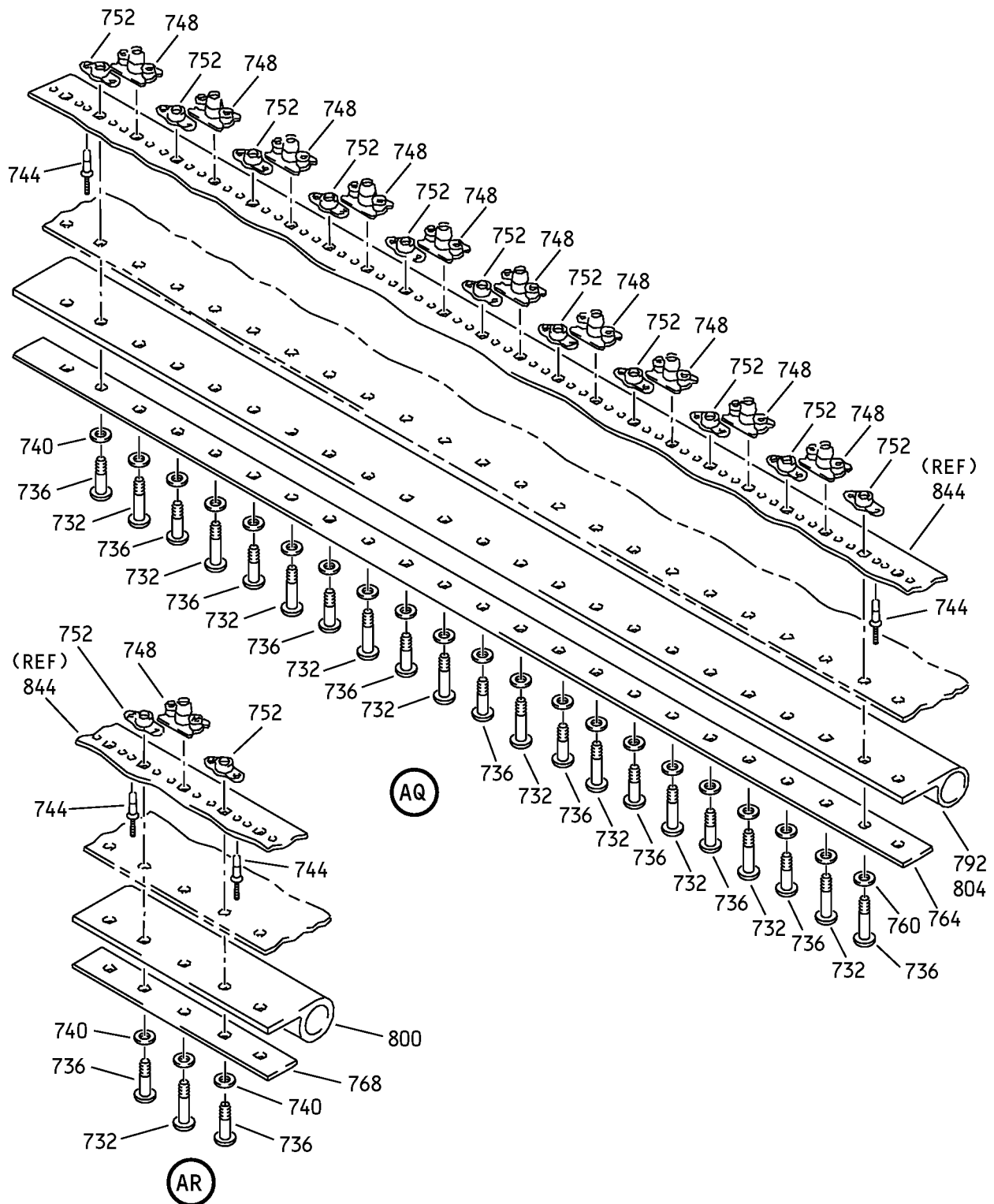
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IPL Figure 4 (Sheet 20 of 29)

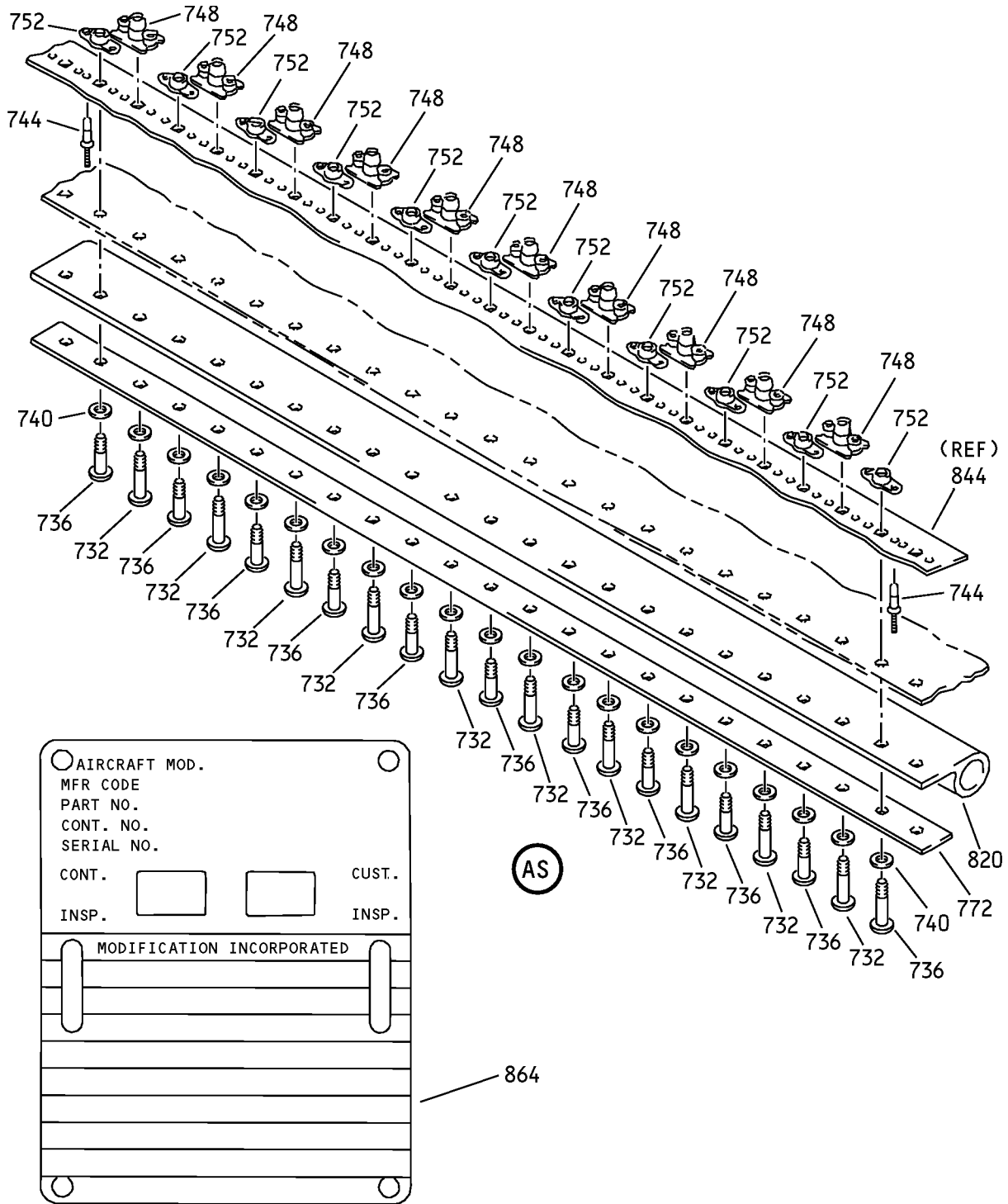
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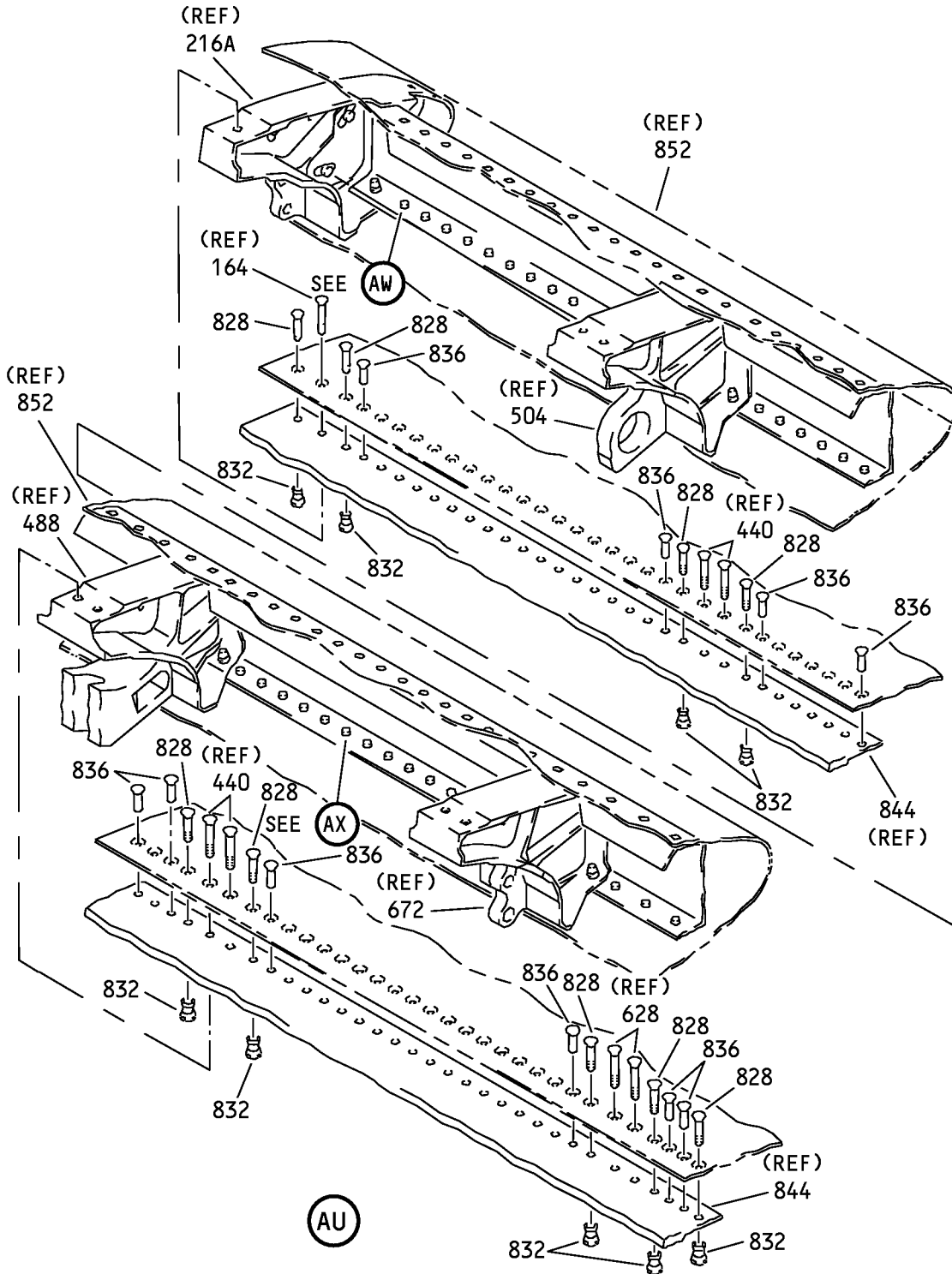
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 22 of 29)

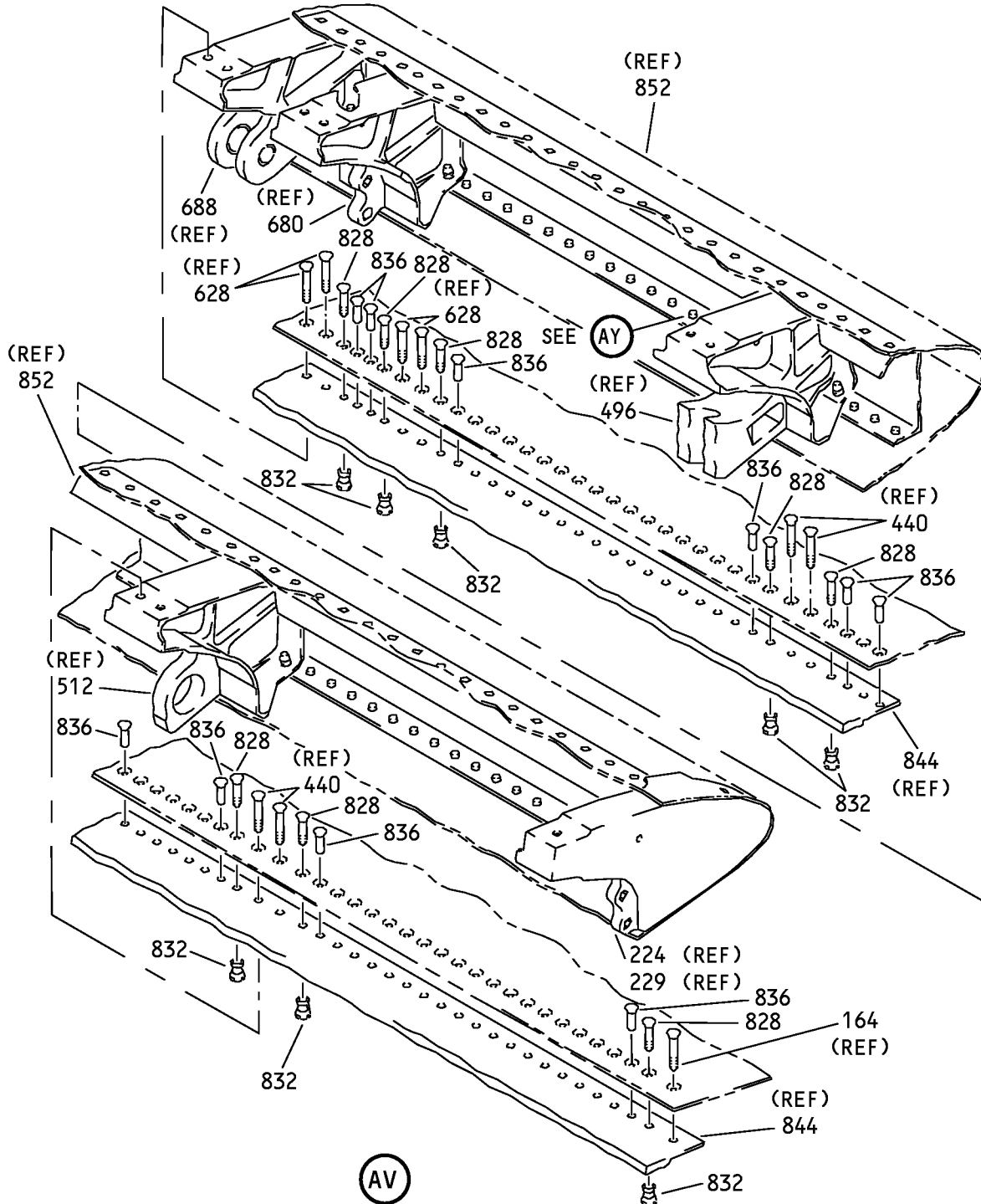
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 23 of 29)

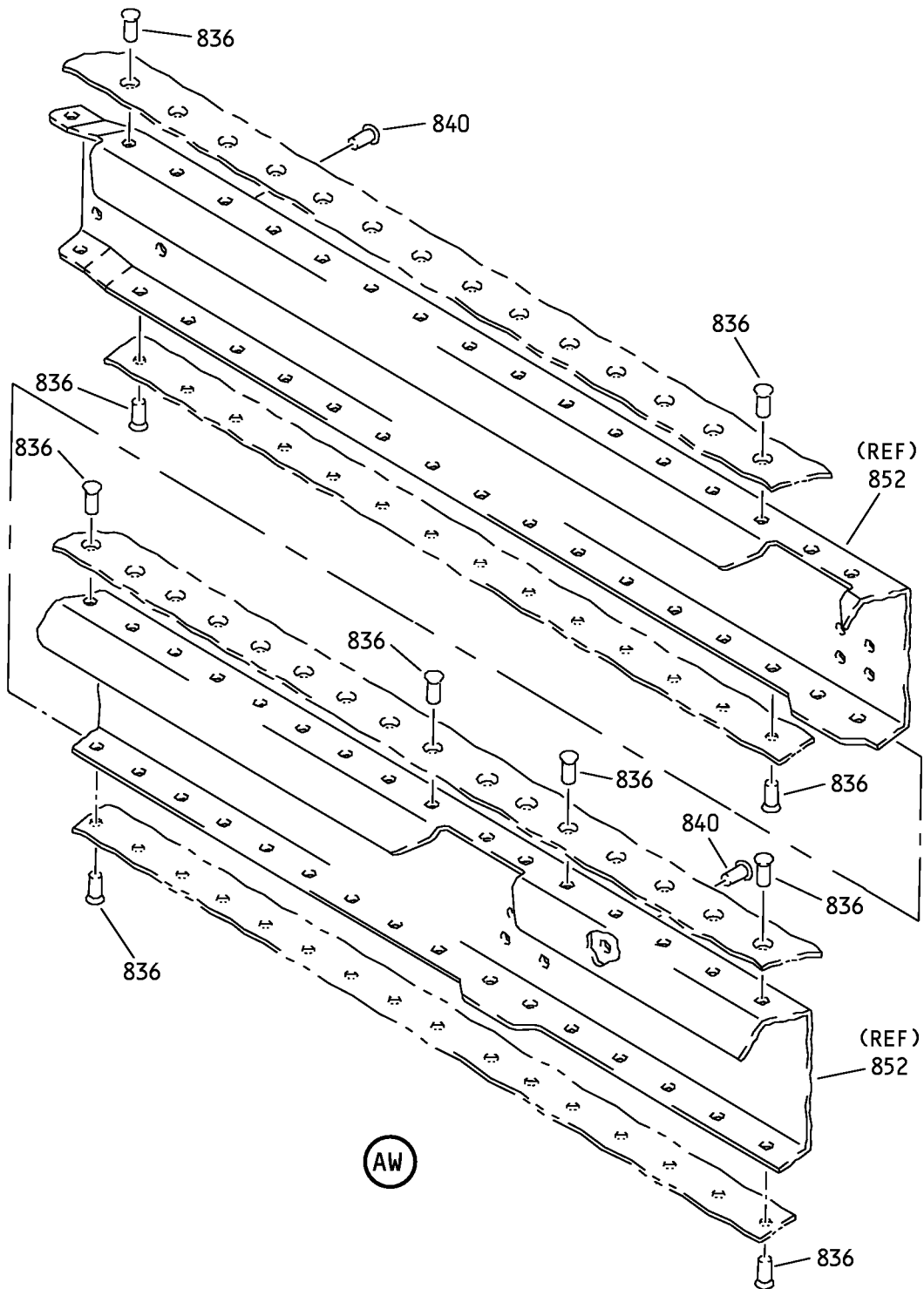
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 24 of 29)

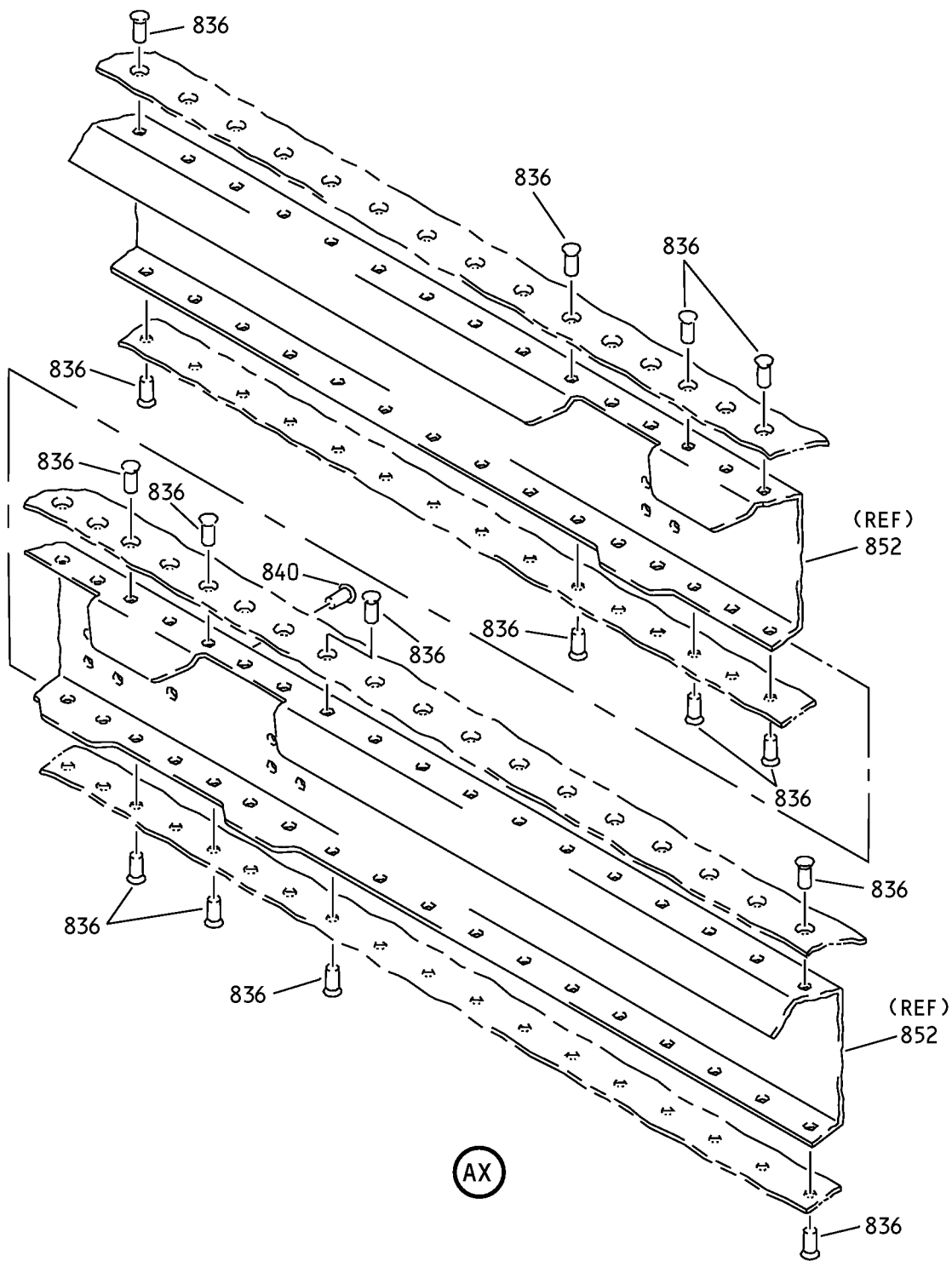
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 25 of 29)

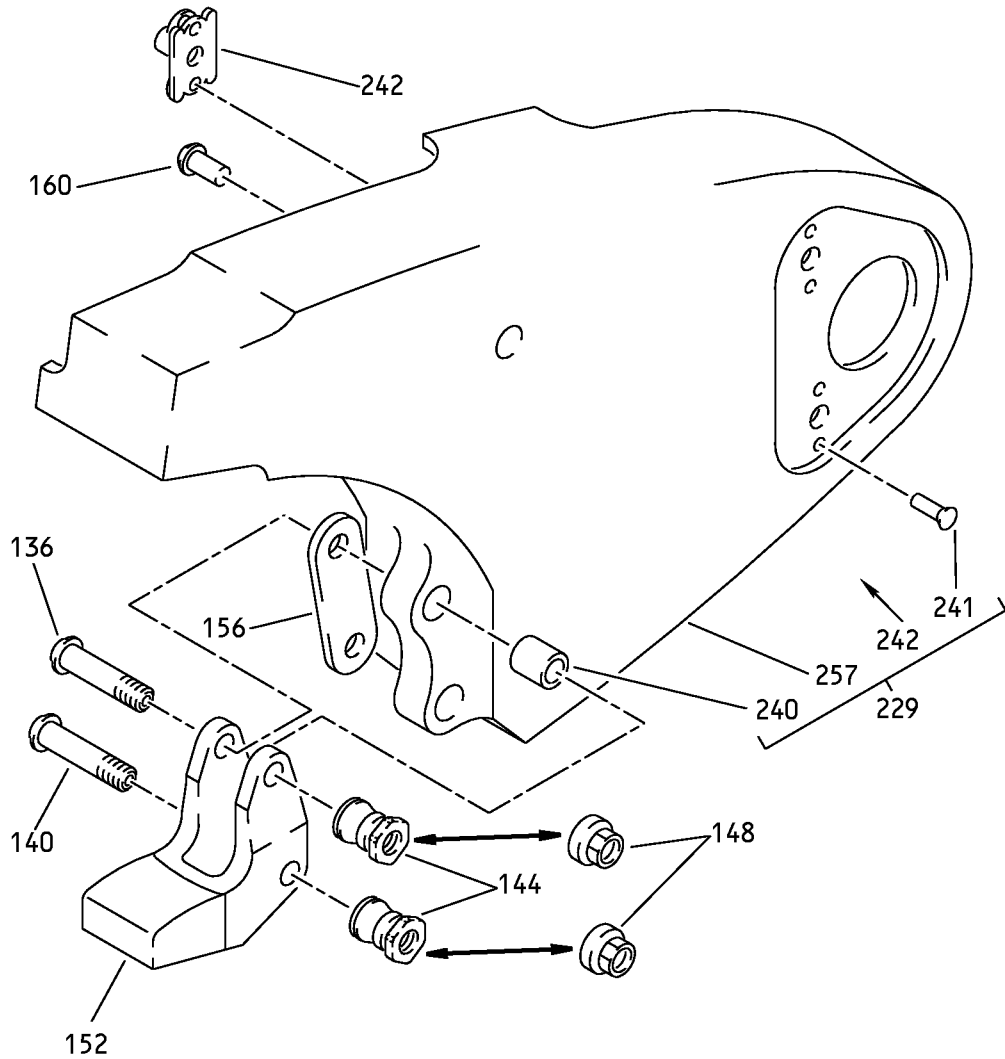
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 26 of 29)

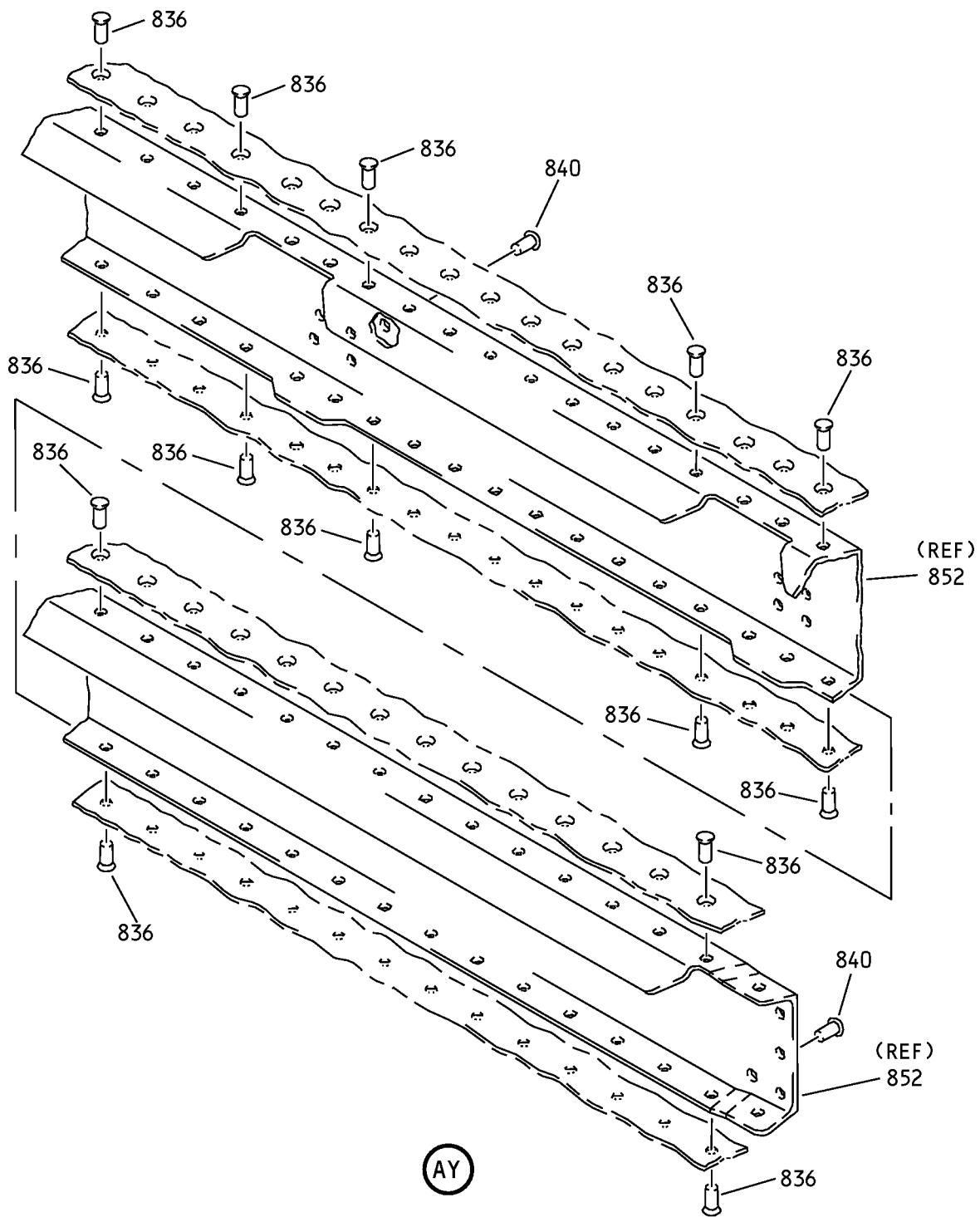
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Wing Leading Edge No. 1 and 8 Slat Assemblies
IPL Figure 4 (Sheet 27 of 29)

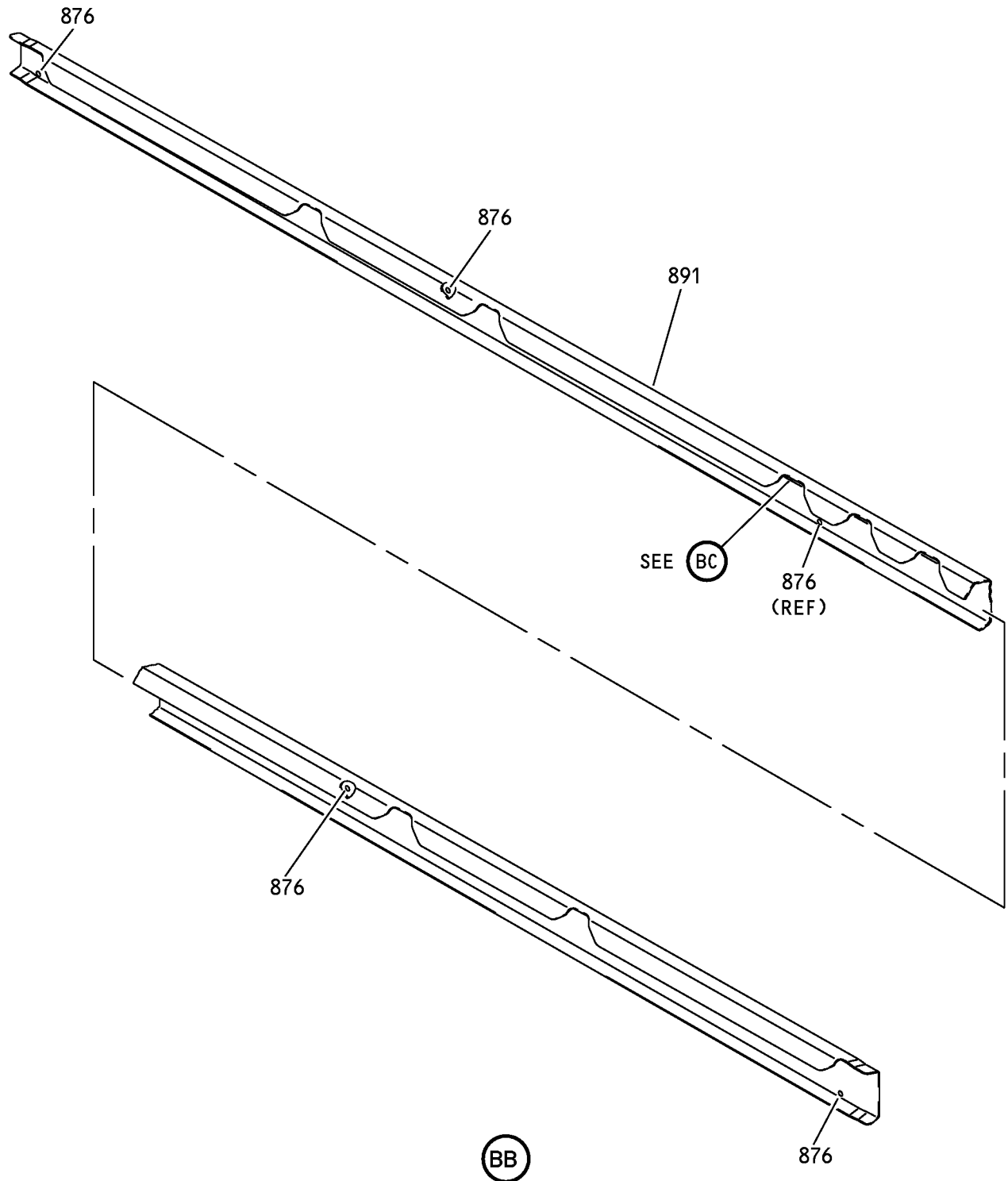
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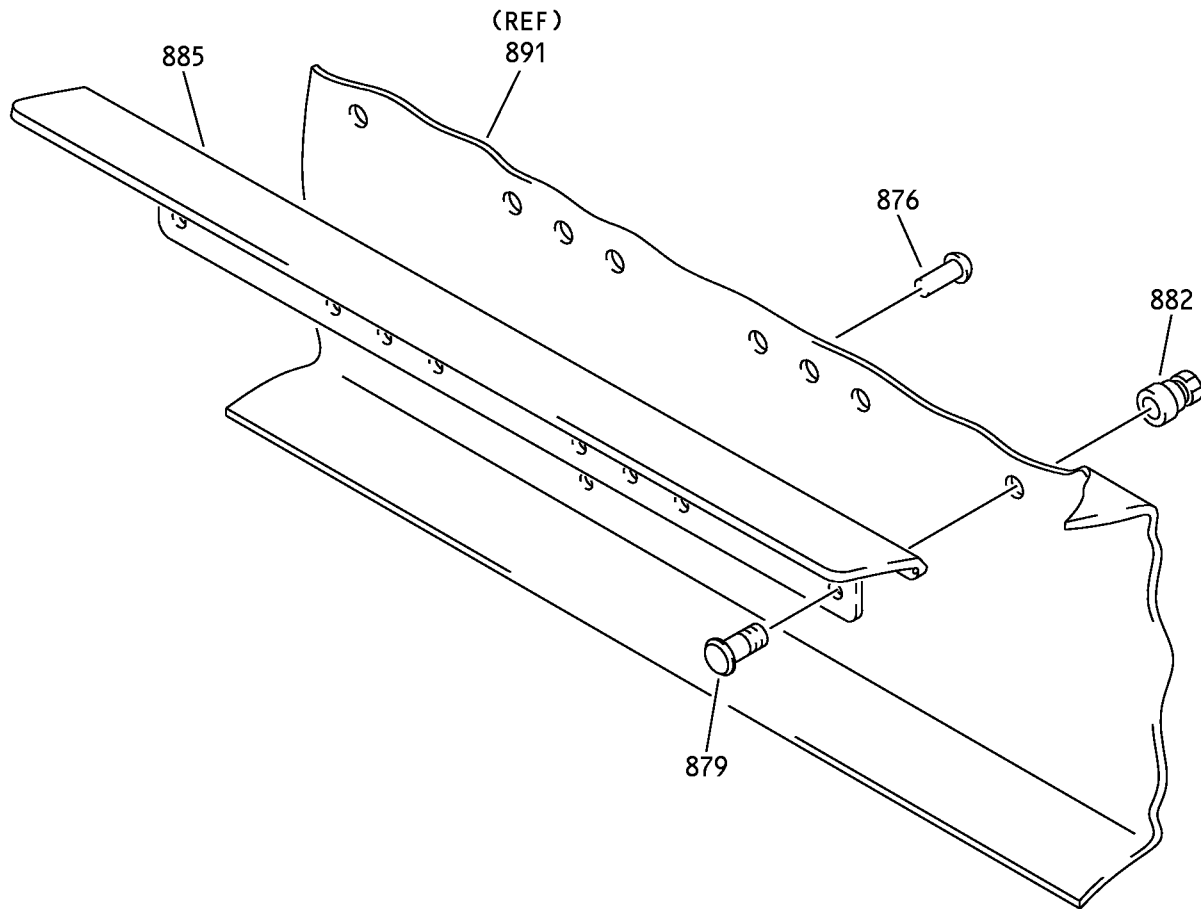
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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Wing Leading Edge No. 1 and 8 Slat Assemblies
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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-1A	114A5010-3									C	RF
-1B	114A5010-201									G	RF
-1C	114A5010-11									J	RF
-4	114A5010-4									D	RF
-4A	114A5010-202									H	RF
-4B	114A5010-12									K	RF
8	BACB30NR8K22									C, D, G-K	2
12	BACW10BP8ACU									C, D, G-K	2
-16	BACW10BP8APU										
16A	BACW10BP8DP									C, D, G-K	2
-20	H01-8BAC										
20A	H51650-8BAC									C, D, G-K	2
24	114A7511-5									C, D	1
-24A	114A7511-201									G, H	1
-24B	114A7511-7									J, K	1
-24C	114A7511-5									J, K	1
28	114A7512-5									C, D	1
-28A	114A7512-201									G, H	1
-28B	114A7512-7									J, K	1
-28C	114A7512-5									J, K	1
32	BACB28AT11B018C									C, D, G-K	1
36	BACB28AP08P018									C, D, G-K	1
40	114A7511-6									C, D, J, K	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-40A	114A7511-202		. .	TRACK						G, H	1
				(USED ON ITEM 24A)							
-40B	114A7511-8		. .	TRACK						J, K	1
				(USED ON ITEM 24B)							
44	114A7512-6		. .	TRACK						C, D, J, K	1
				(USED ON ITEMS 28, 28C)							
-44A	114A7512-202		. .	TRACK						G, H	1
				(USED ON ITEM 28A)							
-44B	114A7512-8		. .	TRACK						J, K	1
				(USED ON ITEM 28B)							
48	BACB28AK08-034		. .	BUSHING						C, D, G-K	2
52	114A5010-5		. .	SUB ASSY-SLAT BOX						C	1
-52A	114A5010-203		. .	SUB ASSY-SLAT BOX						G	1
-52B	114A5010-13		. .	SUB ASSY-SLAT BOX						J	1
-56	114A5010-6		. .	SUB ASSY-SLAT BOX						D	1
-56A	114A5010-204		. .	SUB ASSY-SLAT BOX						H	1
-56B	114A5010-14		. .	SUB ASSY-SLAT BOX						K	1
60	BACB30NT3K7		. .	BOLT						C, D, G-K	4
64	NAS1149D0316J		. .	WASHER						C, D, G-K	4
68	BACR15GE3CW		. .	RIVET						C, D, G, H	8
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 68A)							
-68A	AF5141-3C		. .	RIVET						C, D, G, H	8
				(V53551)							
				(SPEC BACR15DR3AC)							
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 68)							
-72	MF51594-3-2BAC			DELETED							
72A	MF51594-3-4BAC		. .	NUTPLATE						C, D, G-K	4
				(V15653)							
				(SPEC BACN10YF34CD)							
76	114A5102-11		. .	RETAINER-BOLT						C, D	2
				(OPT ITEM 76A)							
-76A	114A5102-19		. .	RETAINER-BOLT						C, D	2
				(OPT ITEM 76)							
-76B	114A5102-19		. .	RETAINER-BOLT						G-K	2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
80	HST11AG6-5		. .	BOLT						C, D, G, H	8
				(V06725)							
				(SPEC BACB30VU6K5)							
				(OPT HST11AG6-5 (V73197))							
				(OPT HST11AG6-5 (V56878))							
				(OPT HST11AG6-5 (V0PTK6))							
-80A	WC331K6-5		. .	BOLT						J, K	8
				(V60516)							
				(SPEC BACB30YP6K5)							
-84	NAS1149D0332J			DELETED							
84A	NAS1149D0363J		. .	WASHER						C, D, G-K	8
88	H52732-3CD		. .	NUT						C, D, G-K	8
				(V15653)							
				(SPEC BACN10YR3CD)							
				(OPT PLH53CD (V62554))							
92	114A9202-1		. .	TAB-FLEX						C, G, J	2
				(OPT ITEM 92A)							
-92A	114A9202-7		. .	TAB-FLEX						C, G, J	2
				(OPT ITEM 92)							
-96	114A9202-2		. .	TAB-FLEX						D, H, K	2
				(OPT ITEM 96A)							
-96A	114A9202-8		. .	TAB-FLEX						D, H, K	2
				(OPT ITEM 96)							
100	BACB30MR4K16		. .	BOLT						C, D	3
-100A	BACB30MR4A16		. .	BOLT						G-K	3
104	NAS42DD8A54FC		. .	SPACER						C, D, G-K	3
108	69-77559-1		. .	WASHER-ALUMINUM						C, D, G-K	3
112	69-77559-3		. .	WASHER-ALUMINUM						C, D, G-K	2
116	69-77600-1		. .	SLEEVE						C, D, G-K	3
				(OPT ITEM 116A)							
-116A	114A9001-1		. .	SLEEVE						C, D, G-K	3
				(OPT ITEM 116)							
120	69-77562-1		. .	RETAINER-SPR						C, D, G-K	3
124	114W4709-1		. .	SPRING						C, D, G-K	3
128	114A9011-1		. .	SEAL-END, OUTBD						C, G, J	1
-132	114A9011-2		. .	SEAL-END, OUTBD						D, H, K	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4- 136	HLT422AP6-10		. .	BOLT							C, D, G-K	2
				(V97928)								
				(SPEC BACB30YK6K10)								
				(OPT HLT422AP6-10 (V06725))								
				(OPT HLT422AP6-10 (V60516))								
140	HLT422AP6-11		. .	BOLT							C, D, G-K	2
				(V97928)								
				(SPEC BACB30YK6K11)								
				(OPT HLT422AP6-11 (V06725))								
				(OPT HLT422AP6-11 (V60516))								
144	BACC30CP6S		. .	COLLAR							C, D, J, K	4
				(OPT ITEM 148)								
-144A	BACC30CP6S		. .	COLLAR							G, H	4
				(OPT ITEM 148A)								
148	BACN10YT3CS		. .	NUT							C, D, J, K	4
				(OPT ITEM 144)								
-148A	BACN10YT3CSA		. .	NUT							G, H	4
				(OPT ITEM 144A)								
152	114A7701-1		. .	FITTING-END STOP							C, D, G-K	2
156	114A5107-1		. .	SHIM-END STOP							C, D, G-K	2
160	BACR15FT8D		. .	RIVET							C, D, G-K	2
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 160A)								
-160A	BACR15FT8AD		. .	RIVET							C, D, G-K	2
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 160)								
164	HST11AG6-5		. .	BOLT							C, D, G-K	2
				(V06725)								
				(SPEC BACB30VU6K5)								
				(OPT HST11AG6-5 (V73197))								
				(OPT HST11AG6-5 (V56878))								
				(OPT HST11AG6-5 (V0PTK6))								
168	HST11AG6-4		. .	BOLT							C, D, G-K	4
				(V06725)								
				(SPEC BACB30VU6K4)								
				(OPT HST11AG6-4 (V73197))								
				(OPT HST11AG6-4 (V0PTK6))								
				(OPT HST11AG6-4 (V56878))								

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4- 172	HST11AG6-3		.	.	BOLT						C, D, G-K	4
					(V06725)							
					(SPEC BACB30VU6K3)							
					(OPT HST11AG6-3 (V73197))							
					(OPT HST11AG6-3 (V56878))							
					(OPT HST11AG6-3 (V0PTK6))							
176	HST10AG6-3		.	.	BOLT						C, D, G-K	6
					(V0PTK6)							
					(SPEC BACB30VT6K3)							
					(OPT HST10AG6-3 (V06725))							
					(OPT HST10AG6-3 (V56878))							
					(OPT HST10AG6-3 (V73197))							
180	HST79CY6		.	.	COLLAR						C, D, G-K	16
					(V73197)							
					(SPEC BACC30BL6)							
					(OPT HST79-6 (V92215))							
					(OPT HST79CY6 (V56878))							
					(OPT HST79CY6 (V5M902))							
184	NAS1399D5A2		.	.	RIVET						C, D, G-K	12
188	BACR15GF6D		.	.	RIVET						C, D, G-K	7
					(SIZE DETERMINED ON INST)							
192	BACB30NT3K7		.	.	BOLT						C, D, G-K	11
196	NAS1149D0316J		.	.	WASHER						C, D, G-K	11
200	BACR15GE3CW		.	.	RIVET						C, D, G-K	22
					(SIZE DETERMINED ON INST)							
					(OPT ITEM 200A)							
-200A	AF5141-3C		.	.	RIVET						C, D, G-K	22
					(V53551)							
					(SPEC BACR15DR3AC)							
					(SIZE DETERMINED ON INST)							
					(OPT ITEM 200)							
204	BACN10JP3ACD		.	.	NUTPLATE						C, D, G-K	2
					(OPT ITEM 204A, 204C)							
-204A	BACN10YF31CD		.	.	NUTPLATE						C, D, G-K	2
					(OPT ITEM 204, 204B)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-204B	MF53050-3CD		. .	NUTPLATE						C, D, G-K	2
				(V15653)							
				(SPEC BACN10JN3CD)							
				(OPT BRFM20C3D (V52828))							
				(OPT 102F9201M3 (V72962))							
				(OPT NS202487-02 (V80539))							
				(OPT MF51637-3 (V15653))							
				(OPT ITEM 204, 204A)							
208	MF51594-3-3BAC		. .	NUTPLATE						C, D, G-K	2
				(V15653)							
				(SPEC BACN10YF33CD)							
212	MF51594-3-4BAC		. .	NUTPLATE						C, D, G-K	7
				(V15653)							
				(SPEC BACN10YF34CD)							
-216	114A7411-1			DELETED							
216A	114A7411-5		. .	RIB ASSY-OUTBD END SS 517.77						C, G, J	1
-220	114A7411-2			DELETED							
-220A	114A7411-6		. .	RIB ASSY-OUTBD END SS 517.77						D, H, K	1
224	114A7412-1		. .	RIB ASSY-INBD END SS 384.97						C	1
-228	114A7412-2		. .	RIB ASSY-INBD END SS 384.97						D	1
229	114A7412-201		. .	RIB ASSY-INBD END SS 384.97						G, J	1
-230	114A7412-202		. .	RIB ASSY-INBD END SS 384.97						H, K	1
232	BACR15BA3AD		. . .	RIVET						C, D, G-K	6
				(SIZE DETERMINED ON INST)							
				(USED ON ITEMS 216A, 220A)							
236	BACN10JP4ACD		. . .	NUTPLATE						C, D, G-K	3
				(USED ON ITEMS 216A, 220A)							
240	BACB28U3C030		. . .	BUSHING						C, D, G-K	2
241	BACR15BA3ADC		. . .	RIVET						G-K	4
				(SIZE DETERMINED ON INST)							
				(USED ON ITEMS 229, 230)							
242	BACN10YF31CD		. . .	NUTPLATE						G-K	2
				(USED ON ITEMS 229, 230)							
-244	114A7411-3			DELETED							
244A	114A7411-7		. . .	RIB						C, G, J	1
				(USED ON ITEM 216A)							
-248	114A7411-4			DELETED							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-248A	114A7411-8		. . .	RIB						D, H, K	1
				(USED ON ITEM 220A)							
252	114A7412-3		. . .	RIB						C	1
				(USED ON ITEM 224)							
-256	114A7412-4		. . .	RIB						D	1
				(USED ON ITEM 228)							
257	114A7412-203		. . .	RIB						G, J	1
				(USED ON ITEM 229)							
-258	114A7412-204		. . .	RIB						H, K	1
				(USED ON ITEM 230)							
260	BACS40R009C010F		. .	SHIM						C, D, G-K	AR
264	BACS40R008B009F		. .	SHIM						C, D, G-K	AR
268	BACS40R010C011F		. .	SHIM						C, D, G-K	AR
272	114A9104-1		. .	SEAL-SPONGE						C, D, G-K	2
276	BACB30NT3K3		. .	BOLT						C, D, G-K	6
280	BACB30NT3K7		. .	BOLT						C, D, G-K	2
284	NAS1149D0316J		. .	WASHER						C, D, G-K	8
288	BACR15GE3CW		. .	RIVET						C, D, G-K	16
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 288A)							
-288A	AF5141-3C		. .	RIVET						C, D, G-K	16
				(V53551)							
				(SPEC BACR15DR3AC)							
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 288)							
292	BACN10JP3ACD		. .	NUTPLATE						C, D, G-K	6
				(OPT ITEM 292A, 292B)							
-292A	BACN10YF31CD		. .	NUTPLATE						C, D, G-K	6
				(OPT ITEM 292, 292B)							
-292B	MF53050-3CD		. .	NUTPLATE						C, D, G-K	6
				(V15653)							
				(SPEC BACN10JN3CD)							
				(OPT BRFM20C3D (V52828))							
				(OPT 102F9201M3 (V72962))							
				(OPT NS202487-02 (V80539))							
				(OPT MF51637-3 (V15653))							
				(OPT ITEM 292, 292A)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
296	MF51594-3-4BAC		.	.	NUTPLATE (V15653) (SPEC BACN10YF34CD)					C, D, G-K	1
300	MF51594-3-3BAC		.	.	NUTPLATE (V15653) (SPEC BACN10YF33CD)					C, D, G-K	1
304	114A9102-22		.	.	RETAINER-SEAL					C, D, G-K	1
308	114A9102-23		.	.	RETAINER-SEAL					C, D, G-K	1
-312	10-60754-1143				DELETED						
312A	BA202179		.	.	SEAL-BULB (VU1901) (SPEC 10-60754-1143)					C, G, J	1
-316	10-60754-1144				DELETED						
-316A	BA202180		.	.	SEAL-BULB (VU1901) (SPEC 10-60754-1144)					D, H, K	1
-320	10-60754-1145				DELETED						
320A	BA202181		.	.	SEAL-BULB (VU1901) (SPEC 10-60754-1145)					C, G, J	1
-324	10-60754-1146				DELETED						
-324A	BA202182		.	.	SEAL-BULB (VU1901) (SPEC 10-60754-1146)					D, H, K	1
328	BACR15FT5AD		.	.	RIVET (SIZE DETERMINED ON INST)					C, D, G-K	4
332	114A5101-3		.	.	TARGET ASSY-OUTBD					C, G, J	1
-336	114A5101-4		.	.	TARGET ASSY-OUTBD					D, H, K	1
340	114A5101-5		.	.	TARGET ASSY-INBD					C, G, J	1
-344	114A5101-6		.	.	TARGET ASSY-INBD					D, H, K	1
-348	BACS12GX08H7				DELETED						
348A	BACB30LH2K3		.	.	BOLT					C, D, G-K	2
352	NAS1149DN816J		.	.	WASHER					C, D, G-K	2
356	H52732-08CD		.	.	NUT (V15653) (SPEC BACN10YR08CD) (OPT PLH508CD (V62554))					C, D, G-K	2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
360	114A5108-2		. . .	BRACKET-TARGET						C, G, J	1
				(USED ON ITEM 332)							
-364	114A5108-1		. . .	BRACKET-TARGET						D, H, K	1
				(USED ON ITEM 336)							
368	114A5108-3		. . .	BRACKET-TARGET						C, G, J	1
				(USED ON ITEM 340)							
-372	114A5108-4		. . .	BRACKET-TARGET						D, H, K	1
				(USED ON ITEM 344)							
376	114A5107-4		. . .	SHIM-LAMINATED						C, D, G-K	1
380	114A5103-1		. . .	TARGET-SENSOR						C, D, G-K	1
384	BACB30VF3K3		. .	BOLT						C, D, G-K	8
388	BACB30NT3K4		. .	BOLT						C, D, G-K	8
-392	NAS1149D0332J			DELETED							
392A	NAS1149D0363J		. .	WASHER						C, D, G-K	16
396	BACW10P42S		. .	WASHER						C, D, G-K	8
400	H52732-3CD		. .	NUT						C, D, G-K	16
				(V15653)							
				(SPEC BACN10YR3CD)							
				(OPT PLH53CD (V62554))							
404	114A5104-3		. .	TAB-SKIN						C, G, J	1
-408	114A5104-4		. .	TAB-SKIN						D, H, K	1
412	114A5104-1		. .	TAB-SKIN						C, G, J	1
-416	114A5104-2		. .	TAB-SKIN						D, H, K	1
420	114A5105-12		. .	SUPPORT-SKIN TAB						C, D, G-K	1
424	114A5105-11		. .	SUPPORT-SKIN TAB						C, D, G-K	1
428	114A5105-10		. .	SUPPORT-SKIN TAB						C, D, G-K	1
432	114A5105-9		. .	SUPPORT-SKIN TAB						C, D, G-K	1
436	BACR15FT8D		. .	RIVET						C, D, G-K	4
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 436A)							
-436A	BACR15FT8AD		. .	RIVET						C, D, G-K	4
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 436)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4- 440	HST11AG6-5		. .	BOLT							C, D, G-K	8
				(V06725)								
				(SPEC BACB30VU6K5)								
				(OPT HST11AG6-5 (V73197))								
				(OPT HST11AG6-5 (V56878))								
				(OPT HST11AG6-5 (V0PTK6))								
444	HST11AG6-4		. .	BOLT							C, D, G-K	26
				(V06725)								
				(SPEC BACB30VU6K4)								
				(OPT HST11AG6-4 (V73197))								
				(OPT HST11AG6-4 (V0PTK6))								
				(OPT HST11AG6-4 (V56878))								
448	HST10AG6-3		. .	BOLT							C, D, G-K	24
				(V0PTK6)								
				(SPEC BACB30VT6K3)								
				(OPT HST10AG6-3 (V06725))								
				(OPT HST10AG6-3 (V56878))								
				(OPT HST10AG6-3 (V73197))								
452	HST79CY6		. .	COLLAR							C, D, G-K	56
				(V73197)								
				(SPEC BACC30BL6)								
				(OPT HST79-6 (V92215))								
				(OPT HST79CY6 (V56878))								
				(OPT HST79CY6 (V5M902))								
456	BACR15GF6D		. .	RIVET							C, D, G-K	30
				(SIZE DETERMINED ON INST)								
460	BACB30NT3K7		. .	BOLT							C, D, G-K	44
464	NAS1149D0316J		. .	WASHER							C, D, G-K	44
468	BACR15GE3CW		. .	RIVET							C, D, G-K	88
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 468A)								
-468A	AF5141-3C		. .	RIVET							C, D, G-K	88
				(V53551)								
				(SPEC BACR15DR3AC)								
				(SIZE DETERMINED ON INST)								
				(OPT ITEM 468)								
472	BACN10JP3ACD		. .	NUTPLATE							C, D, G-K	8
				(OPT ITEM 472A, 472B)								
-472A	BACN10YF31CD		. .	NUTPLATE							C, D, G-K	8
				(OPT ITEM 472, 472B)								

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4- -472B	MF53050-3CD		.	.	NUTPLATE (V15653) (SPEC BACN10JN3CD) (OPT BRFM20C3D (V52828)) (OPT 102F9201M3 (V72962)) (OPT NS202487-02 (V80539)) (OPT MF51637-3 (V15653)) (OPT ITEM 472, 472A)						C, D, G-K	8
476	MF51594-3-3BAC		.	.	NUTPLATE (V15653) (SPEC BACN10YF33CD)						C, D, G-K	27
480	MF51594-3-4BAC		.	.	NUTPLATE (V15653) (SPEC BACN10YF34CD)						C, D, G-K	7
484	MF51594-3-2BAC		.	.	NUTPLATE (V15653) (SPEC BACN10YF32CD)						C, D, G-K	2
488	114A7111-1		.	.	RIB ASSY-OUTBD AUX ARM SS 480.97 (FOR DETAILS SEE FIG. 2)						C, G, J	1
-492	114A7111-2		.	.	RIB ASSY-OUTBD AUX ARM SS 480.97 (FOR DETAILS SEE FIG. 2)						D, H, K	1
496	114A7112-1		.	.	RIB ASSY-INBD AUX ARM SS 422.97 (FOR DETAILS SEE FIG. 2)						C, G, J	1
-500	114A7112-2		.	.	RIB ASSY-INBD AUX ARM SS 422.97 (FOR DETAILS SEE FIG. 2)						D, H, K	1
504	114A7011-1		.	.	RIB ASSY-OUTBD MAIN TRACK SS 494.85						C, G, J	1
-508	114A7011-2		.	.	RIB ASSY-OUTBD MAIN TRACK SS 494.85						D, H, K	1
512	114A7012-1		.	.	RIB ASSY-INBD MAIN TRACK SS 409.09						C, J	1
-512A	114A7012-201		.	.	RIB ASSY-INBD MAIN TRACK SS 409.09						G	1
-516	114A7012-2		.	.	RIB ASSY-INBD MAIN TRACK SS 409.09						D, K	1
-516A	114A7012-202		.	.	RIB ASSY-INBD MAIN TRACK SS 409.09						H	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
520	ADW08V301NC		. . .	BEARING						C, D, G-K	1
				(V15860)							
				(SPEC BACB10FA08GC)							
				(OPT SWKRS08-350SC (V81376))							
				(OPT KSC152200BZ08GC							
				(V50632))							
				(OPT KWDB08-35 (V97613))							
				(OPT WES08FAGC (V73134))							
				(OPT WHTFA08VC (VS0352))							
524	114A7011-3		. . .	RIB						C, G, J	1
				(USED ON ITEM 504)							
-528	114A7011-4		. . .	RIB						D, H, K	1
				(USED ON ITEM 508)							
532	114A7012-3		. . .	RIB						C, J	1
				(USED ON ITEM 512)							
-532A	114A7012-203		. . .	RIB						G	1
				(USED ON ITEM 512A)							
-536	114A7012-4		. . .	RIB						D, K	1
				(USED ON ITEM 516)							
-536A	114A7012-204		. . .	RIB						H	1
				(USED ON ITEM 516A)							
540	BACS40R010C017F		. .	SHIM						C, D, G-K	AR
544	BACS40R009B017F		. .	SHIM						C, D, G-K	AR
548	BACS40R008B018F		. .	SHIM						C, D, G-K	AR
552	BACS40R008B020F		. .	SHIM						C, D, G-K	AR
556	BACS40R010C020F		. .	SHIM						C, D, G-K	AR
560	BACS40R010C030F		. .	SHIM						C, D, G-K	AR
564	HLT422AP8-13		. .	BOLT						C, D, G-K	2
				(V97928)							
				(SPEC BACB30YK8K13)							
				(OPT HLT422AP8-13 (V06725))							
				(OPT HLT422AP8-13 (V60516))							
568	HLT422AP10-16		. .	BOLT						C, D, G-K	2
				(V97928)							
				(SPEC BACB30YK10K16)							
				(OPT HLT422AP10-16 (V06725))							
				(OPT HLT422AP10-16 (V60516))							
572	BACC30CP8S		. .	COLLAR						C, D, J, K	2
				(OPT ITEM 576)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-572A	BACC30CP8S		. .	COLLAR						G, H	2
				(OPT ITEM 576A)							
576	BACN10YT4CS		. .	NUT						C, D, J, K	2
				(OPT ITEM 572)							
-576A	BACN10YT4CSA		. .	NUT						G, H	2
				(OPT ITEM 572A)							
580	BACC30CP10S		. .	COLLAR						C, D, J, K	2
				(OPT ITEM 584)							
-580A	BACC30CP10S		. .	COLLAR						G, H	2
				(OPT ITEM 584A)							
584	BACN10YT5CS		. .	NUT						C, D, J, K	2
				(OPT ITEM 580)							
-584A	BACN10YT5CSA		. .	NUT						G, H	2
				(OPT ITEM 580A)							
588	114A5107-2		. .	SHIM-UPSTOP						C, D, G-K	2
592	114A7702-1		. .	FITTING-UPSTOP						C, D, G-K	2
596	BACB30NT3K7		. .	BOLT						C, D, G-K	2
600	NAS1149D0316J		. .	WASHER						C, D, G-K	2
604	BACR15GE3CW		. .	RIVET						C, D, G-K	4
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 604A)							
-604A	AF5141-3C		. .	RIVET						C, D, G-K	4
				(V53551)							
				(SPEC BACR15DR3AC)							
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 604)							
608	MF51594-3-2BAC		. .	NUTPLATE						C, D, G-K	1
				(V15653)							
				(SPEC BACN10YF32CD)							
612	MF51594-3-4BAC		. .	NUTPLATE						C, D, G-K	1
				(V15653)							
				(SPEC BACN10YF34CD)							
616	114A5102-17		. .	RETAINER-BOLT						C, G, J	1
-620	114A5102-18		. .	RETAINER-BOLT						D, H, K	1
624	BACR15FT8D		. .	RIVET						C, D, G-K	3
				(SIZE DETERMINED ON INST)							
				(OPT ITEM 624A)							

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4- -624A	BACR15FT8AD		.	.	RIVET (SIZE DETERMINED ON INST) (OPT ITEM 624)						C, D, G-K	3
628	HST11AG6-5		.	.	BOLT (V06725) (SPEC BACB30VU6K5) (OPT HST11AG6-5 (V73197)) (OPT HST11AG6-5 (V56878)) (OPT HST11AG6-5 (V0PTK6))						C, D, G-K	6
632	HST11AG6-4		.	.	BOLT (V06725) (SPEC BACB30VU6K4) (OPT HST11AG6-4 (V73197)) (OPT HST11AG6-4 (V0PTK6)) (OPT HST11AG6-4 (V56878))						C, D, G-K	20
636	HST10AG6-3		.	.	BOLT (V0PTK6) (SPEC BACB30VT6K3) (OPT HST10AG6-3 (V06725)) (OPT HST10AG6-3 (V56878)) (OPT HST10AG6-3 (V73197))						C, D, G-K	18
640	HST79CY6		.	.	COLLAR (V73197) (SPEC BACC30BL6) (OPT HST79-6 (V92215)) (OPT HST79CY6 (V56878)) (OPT HST79CY6 (V5M902))						C, D, G-K	42
644	BACR15GF6D		.	.	RIVET (SIZE DETERMINED ON INST)						C, D, G-K	20
648	BACB30NT3K7		.	.	BOLT						C, D, G-K	35
652	NAS1149D0316J		.	.	WASHER						C, D, G-K	35
656	BACR15GE3CW		.	.	RIVET (SIZE DETERMINED ON INST) (OPT ITEM 656A)						C, D, G-K	70
-656A	AF5141-3C		.	.	RIVET (V53551) (SPEC BACR15DR3AC) (SIZE DETERMINED ON INST) (OPT ITEM 656)						C, D, G-K	70
660	BACN10JP3ACD		.	.	NUTPLATE (OPT ITEM 660A, 660B)						C, D, G-K	6

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-660A	BACN10YF31CD		. .							C, D, G-K	6
-660B	MF53050-3CD		. .							C, D, G-K	6
664	MF51594-3-2BAC		. .							C, D, G-K	12
668	MF51594-3-3BAC		. .							C, D, G-K	17
672	114A7311-1		. .							C, J	1
-672A	114A7311-201		. .							G	1
-676	114A7311-2		. .							D, K	1
-676A	114A7311-202		. .							H	1
680	114A7312-1		. .							C, J	1
-680A	114A7312-201		. .							G	1
-684	114A7312-2		. .							D, K	1
-684A	114A7312-202		. .							H	1
688	114A7211-1		. .							C, J	1
-688A	114A7211-201		. .							G	1
-692	114A7211-2		. .							D, K	1
-692A	114A7211-202		. .							H	1
696	BACB28AP10P032		. . .							C, D, G-K	1
700	BACB28AT14B032C		. . .							C, D, G-K	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY	
			1	2	3	4	5	6	7			
4-												
704	114A7211-3									RIB	C, J	1
-704A	114A7211-203									RIB	G	1
-708	114A7211-4									RIB	D, K	1
-708A	114A7211-204									RIB	H	1
712	BACS40R010C017F									SHIM	C, D, G-K	AR
716	BACS40R009B017F									SHIM	C, D, G-K	AR
720	BACS40R008B020F									SHIM	C, D, G-K	AR
724	BACS40R008B018F									SHIM	C, D, G-K	AR
728	BACS40R010C020F									SHIM	C, D, G-K	AR
732	BACB30NT3K7									BOLT	C, D, G-K	52
736	BACB30NT3K3									BOLT	C, D, G-K	59
740	NAS1149D0316J									WASHER	C, D, G-K	111
744	BACR15GE3CW									RIVET (SIZE DETERMINED ON INST) (OPT ITEM 744A)	C, D, G-K	222
-744A	AF5141-3C									RIVET (V53551) (SPEC BACR15DR3AC) (SIZE DETERMINED ON INST) (OPT ITEM 744)	C, D, G-K	222
748	MF51594-3-3BAC									NUTPLATE (V15653) (SPEC BACN10YF33CD)	C, D, G-K	52
752	BACN10JP3ACD									NUTPLATE (OPT ITEM 752A, 752B)	C, D, G-K	59
-752A	BACN10YF31CD									NUTPLATE (OPT ITEM 752, 752B)	C, D, G-K	59
-752B	MF53050-3CD									NUTPLATE (V15653) (SPEC BACN10JN3CD) (OPT BRFM20C3D (V52828)) (OPT 102F9201M3 (V72962)) (OPT NS202487-02 (V80539)) (OPT MF51637-3 (V15653)) (OPT ITEM 752, 752A)	C, D, G-K	59
756	114A9102-30									RETAINER-SEAL	C, D, G-K	1
760	114A9102-2									RETAINER-SEAL	C, D, G-K	2

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY		
			1	2	3	4	5	6	7				
4-													
764	114A9102-3		. .								RETAINER-SEAL	C, D, G-K	2
768	114A9102-4		. .								RETAINER-SEAL	C, D, G-K	2
772	114A9102-5		. .								RETAINER-SEAL	C, D, G-K	1
776	114A9101-45		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 24.36)	C, G, J	1
-780	114A9101-46		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 24.36)	D, H, K	1
784	114A9101-3		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 13.88)	C, G, J	1
-788	114A9101-4		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 13.88)	D, H, K	1
792	114A9101-5		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 23.70)	C, G, J	1
-796	114A9101-6		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-482 X 23.70)	D, H, K	1
800	114A9101-7		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 5.30)	C, D, G-K	2
804	114A9101-9		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 23.70)	C, G, J	1
-808	114A9101-10		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 23.70)	D, H, K	1
812	114A9101-11		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 13.88)	C, G, J	1
-816	114A9101-12		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 13.88)	D, H, K	1
820	114A9101-13		. .								SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 24.12)	C, G, J	1

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-824	114A9101-14		. .	SEAL-BULB (MAKE FROM EXTR RUBBER 10-60754-1129 X 24.12)					D, H, K	1	
828	HST11AG6-3		. .	BOLT (V06725) (SPEC BACB30VU6K3) (OPT HST11AG6-3 (V73197)) (OPT HST11AG6-3 (V56878)) (OPT HST11AG6-3 (V0PTK6))					C, D, G-K	17	
832	HST79CY6		. .	COLLAR (V73197) (SPEC BACC30BL6) (OPT HST79-6 (V92215)) (OPT HST79CY6 (V56878)) (OPT HST79CY6 (V5M902))					C, D, G-K	17	
836	BACR15GF6D		. .	RIVET (SIZE DETERMINED ON INST)					C, D, G-K	263	
840	BACR15BB8AD		. .	RIVET (SIZE DETERMINED ON INST)					C, D, G-K	5	
844	114A6010-1		. .	WEDGE ASSY-BONDED (OPT ITEM 844A, 844B)					C	1	
-844A	114A6010-11		. .	WEDGE ASSY-BONDED (OPT ITEM 844, 844B)					C	1	
-844B	114A6010-21		. .	WEDGE ASSY-BONDED (OPT ITEM 844, 844A)					C	1	
-844C	114A6010-21		. .	WEDGE ASSY-BONDED					G, J	1	
-848	114A6010-2		. .	WEDGE ASSY-BONDED (OPT ITEM 848A, 848B)					D	1	
-848A	114A6010-12		. .	WEDGE ASSY-BONDED (OPT ITEM 848, 848B)					D	1	
-848B	114A6010-22		. .	WEDGE ASSY-BONDED (OPT ITEM 848, 848A)					D	1	
-848C	114A6010-22		. .	WEDGE ASSY-BONDED						1	
852	114A8211-1		. .	BEAM-NOSE					C, J	1	
-856	114A8211-2		. .	BEAM-NOSE					D, K	1	
857	NAS1399D4A2		. .	RIVET					C, D, G-K	4	
858	114A5109-1		. .	VORTILON (OPT ITEM 858A)					C, G, J	1	

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
4-											
-858A	114A5109-7		. .	VORTILON						C, G, J	1
				(OPT ITEM 858)							
-859	114A5109-2		. .	VORTILON						D, H, K	1
				(OPT ITEM 859A)							
-859A	114A5109-8		. .	VORTILON						D, H, K	1
				(OPT ITEM 859)							
860	BACR15BB3AD		. .	RIVET						C, D, G-K	4
				(SIZE DETERMINED ON INST)							
864	MS27253-1		. .	PLATE						C, D, G-K	1
				(OPT ITEM 864A)							
-864A	MS27253F2		. .	PLATE						C, D, G-K	1
				(OPT ITEM 864)							
870	114A8210-201		. .	BEAM ASSY-NOSE						G	1
-873	114A8210-202		. .	BEAM ASSY-NOSE						H	1
876	BACR15BB8AD5C		. . .	RIVET						G, H	5
879	HST10AG6-2		. . .	BOLT						G, H	8
				(V0PTK6)							
				(SPEC BACB30VT6K2)							
				(OPT HST10AG6-2 (V06725))							
				(OPT HST10AG6-2 (V56878))							
				(OPT HST10AG6-2 (V73197))							
882	HST1094KP6		. . .	COLLAR						G, H	8
				(V73197)							
				(SPEC BACC30BS6)							
885	114A8212-201		. . .	BACKUP						G	1
-888	114A8212-202		. . .	BACKUP						H	1
891	114A8211-201		. . .	BEAM						G	1
-894	114A8211-202		. . .	BEAM						H	1
897	BACB30VF3K2		. .	BOLT						G, H	2
900	114A5110-201		. .	PLATE-COVER, END						G	1
-903	114A5110-202		. .	PLATE-COVER, END						H	1

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