



COMPONENT MAINTENANCE MANUAL WITH ILLUSTRATED PARTS LIST

FLIGHT SPOILER ASSEMBLY NO. 5 AND 8

**PART NUMBER
113A4500-1, -2, -3, -4**

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COMPONENT MAINTENANCE MANUAL

Revision No. 8
Jul 01/2009

To: All holders of FLIGHT SPOILER ASSEMBLY NO. 5 AND 8 57-56-56.

Attached is the current revision to this COMPONENT MAINTENANCE MANUAL

The COMPONENT MAINTENANCE MANUAL is furnished either as a printed manual, on microfilm, or digital products, or any combination of the three. This revision replaces all previous microfilm cartridges or digital products. All microfilm and digital products are reissued with all obsolete data deleted and all updated pages added.

For printed manuals, changes are indicated on the List of Effective Pages (LEP). The pages which are revised will be identified on the LEP by an R (Revised), A (Added), O (Overflow, i.e. changes to the document structure and/or page layout), or D (Deleted). Each page in the LEP is identified by Chapter-Section-Subject number, page number and page date.

Pages replaced or made obsolete by this revision should be removed and destroyed.

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TRANSMITTAL LETTER

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113A4500



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Location of Change

Description of Change

NO HIGHLIGHTS

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HIGHLIGHTS

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A = Added, R = Revised, D = Deleted, O = Overflow

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TEMPORARY REVISION AND SERVICE BULLETIN RECORD

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TR AND SB RECORD

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COMPONENT MAINTENANCE MANUAL

INTRODUCTION

1. General

- A. The instructions in this manual supply the data necessary to do the maintenance functions together with the test, fault isolation, repair, and replacement of the defective parts.
- B. This manual is divided into different parts:
 - (1) Title Page
 - (2) Transmittal Letter
 - (3) Highlights
 - (4) List of Effective Pages
 - (5) Table of Contents
 - (6) Temporary Revision & Service Bulletin Record
 - (7) Record of Revisions
 - (8) Record of Temporary Revisions
 - (9) Introduction
 - (10) Procedures & IPL Sections
- C. Components that can be repaired have a different repair number for each specified repair. To find the repair number location of a component, look in the Repair-General procedure at the beginning of the REPAIR section. The Repair-General procedure also has an explanation of the True Position Dimension symbols used.
- D. All dimensions, measures, quantities and weights included are in English units. When metric equivalents are given they will be in the parentheses that follow the English units.
- E. The introduction to the Illustrated Parts List (IPL) shows how the IPL data is used.
- F. Design changes, optional parts, configuration differences and Service Bulletin modifications may cause different part numbers. These part numbers are identified in the IPL with an alphabetical letter which is added to the end of the basic item number. This new item number is referred to as an alpha-variant. Throughout the manual, IPL basic item number references also apply to alpha-variants unless shown differently.
- G. The tool reference numbers found in the individual procedures and in the Special Tools, Fixtures, and Equipment section are used to identify if a tool is a standard tool (STD-XXXX), a commercial tool (COM-XXXX), or a Special Tool (SPL-XXXX). This reference number is also used to distinguish between tools with similar names in the same procedure. These reference numbers are for use in the documentation only. They are not to be used for ordering tools.

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INTRODUCTION

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COMPONENT MAINTENANCE MANUAL

NO. 5 AND 8 FLIGHT SPOILER ASSEMBLY - DESCRIPTION AND OPERATION

1. Description

- A. There are four flight spoiler assemblies, one outboard ground spoiler assembly, and one inboard ground spoiler assembly on each wing. The outboard ground spoiler assemblies are installed on the left and right wings in pair as No. 1 and 12. The spoiler assemblies are in pairs as No. 2 and 11, No. 3 and 10, No. 4 and 9, No. 5 and 8. The inboard ground spoiler assemblies are in pair as No. 6 and 7.
- The flight spoiler assembly has a bonded assembly, two jumper assemblies, a center fitting, and two end ribs. The bonded assembly is made from aluminum honeycomb core reinforced between two aluminum alloy panels. The center fitting and the two end ribs are made from aluminum alloy. The flight spoiler assemblies are attached to the hinges on the rear spar of the wings.

2. Operation

- A. The flight spoiler assemblies are used together with the aileron assemblies for lateral control and speed brakes. The flight spoiler assemblies on the left and right wings can be raised alternately for lateral control during flight, or can be raised together as speed brakes during landing. The flight spoiler assemblies are controlled by actuators.

3. Leading Particulars (Approximate)

- A. Length – 46 inches
- B. Width – 27 inches
- C. Height – 3 inches
- D. Weight – 13 pounds

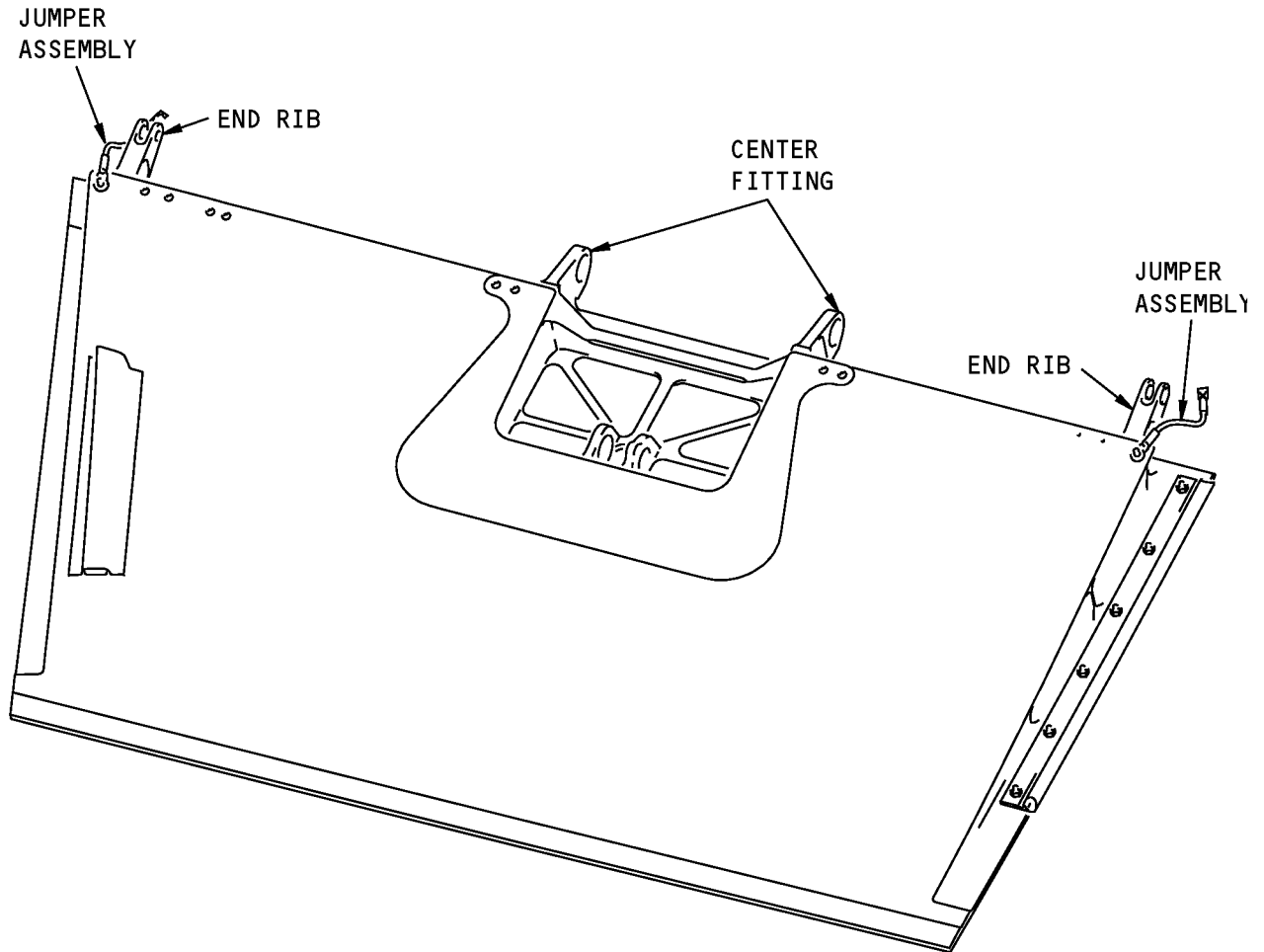
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DESCRIPTION AND OPERATION

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No. 5 and 8 Flight Spoiler Assemblies
Figure 1

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DESCRIPTION AND OPERATION

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TESTING AND FAULT ISOLATION

(NOT APPLICABLE)

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TESTING AND FAULT ISOLATION

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DISASSEMBLY

1. General

- A. This procedure has the data necessary to disassemble the No. 5 and 8 flight spoiler assembly.
- B. Disassemble this component sufficiently to isolate the defects, do the necessary repairs, and put the component back to a serviceable condition.
- C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.

2. Disassembly

- A. Procedure

NOTE: For bearing and bushing replacement, refer to SOPM 20-50-03.

- (1) Use standard industry procedures to disassemble this component.

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DISASSEMBLY

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CLEANING

1. General

- A. This procedure has the data necessary to clean the No. 5 and 8 flight spoiler assembly.
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to IPL Figure 1 for item numbers.

2. Cleaning

- A. Procedure
 - (1) Clean the bearing (20) as specified in SOPM 20-30-01.
 - (2) Use standard industry procedures and refer to SOPM 20-30-03 to clean all parts.

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CLEANING
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CHECK

1. General

- A. This procedure has the data necessary to find defects in the material of the specified parts.
- B. Refer to FITS AND CLEARANCES for the design dimension and wear limits.
- C. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- D. Refer to IPL Figure 1 for item numbers.

2. Check

A. References

Reference	Title
SOPM 20-20-02	PENETRANT METHODS OF INSPECTION
737 NDT Part 1, 51-05-01	Tap Test Inspection of Honeycomb Sandwich Structure
737 NDT Part 4, 51-00-02	Full Depth Honeycomb and Laminate Structure Inspection
737 NDT Part 9, 51-00-01	Non-Destructive Testing

B. Procedure

- (1) Use standard industry procedures to do a visual check of all the parts for defects. Do the penetrant or magnetic particle check if the visual check shows possible damage or if you suspect possible damage on the parts listed below.
 - (a) Do a penetrant check (SOPM 20-20-02) of these parts
 - 1) End rib (165, 170, 175, 180)
 - 2) Center fitting (185, 190)
- (2) Do a check for cracks, corrosion, or missing potting and sealant at the ends of the panel bond assemblies.
- (3) Do a check of the honeycomb structure and bonded parts for evidence of delamination, internal water, scratches, and contour defects.
 - (a) If you see delamination or impact damage when you do a visual check, do an ultrasonic inspection or a tap test to find all damage. Refer to the 737 NDT Part 4, 51-00-02.

NOTE: For the tap test, use a small solid metal disk and tap the surface area lightly but firmly. You will hear a sharp sound when you tap on solid bonded areas, and dull sound on void areas. Refer to the 737 NDT Part 1, 51-05-01.
 - (b) Do a check on areas you suspect of containing water with the radiographic or thermographic method. Refer to the 737 NDT Part 9, 51-00-01.
 - (c) Do a check on the edges of the panel carefully for cuts and abrasions. Delamination starts very easily from damage to an edgemember of honeycomb panel.
- (4) Refer to the applicable 737 Structural Repair Manuals section for allowable damage and repair data.

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CHECK

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REPAIR

1. General

- A. Instructions for repair, refinish, and replacement of the specified subassembly parts are included in each REPAIR when applicable:

PART NUMBER	NAME	REPAIR
—	REFINISH OF OTHER PARTS	1-1
113A4112	END RIB	2-1
113A4500	NO. 5 AND 8 FLIGHT	3-1
113A4510	SPOILER ASSEMBLY	3-2
113A4520	CENTER FITTING	4-1

2. Dimensioning Symbols

- A. Standard True Position Dimensioning Symbols used in the applicable repair procedures are shown in REPAIR-GENERAL, Figure 601.

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REPAIR - GENERAL

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—	STRAIGHTNESS	∅	DIAMETER
▭	FLATNESS	S ∅	SPHERICAL DIAMETER
⊥	PERPENDICULARITY (OR SQUARENESS)	R	RADIUS
//	PARALLELISM	SR	SPHERICAL RADIUS
○	ROUNDNESS	()	REFERENCE
⊘	CYLINDRICITY	BASIC	A THEORETICALLY EXACT DIMENSION USED
⌒	PROFILE OF A LINE	(BSC)	TO DESCRIBE SIZE, SHAPE OR LOCATION OF
⌒	PROFILE OF A SURFACE	OR	A FEATURE. FROM THIS FEATURE PERMISSIBLE
◎	CONCENTRICITY	DIM	VARIATIONS ARE ESTABLISHED BY TOLERANCES ON OTHER DIMENSIONS OR
≡	SYMMETRY		NOTES.
∠	ANGULARITY	-A-	DATUM
↗	RUNOUT	Ⓜ	MAXIMUM MATERIAL CONDITION (MMC)
↗	TOTAL RUNOUT	Ⓛ	LEAST MATERIAL CONDITION (LMC)
⊔	COUNTERBORE OR SPOTFACE	Ⓢ	REGARDLESS OF FEATURE SIZE (RFS)
∇	COUNTERSINK	Ⓟ	PROJECTED TOLERANCE ZONE
⊕	THEORETICAL EXACT POSITION OF A FEATURE (TRUE POSITION)	FIM	FULL INDICATOR MOVEMENT

EXAMPLES

$\boxed{\text{—}} \boxed{0.002}$	STRAIGHT WITHIN 0.002	$\boxed{\text{◎}} \boxed{\text{∅}} \boxed{0.0005} \boxed{\text{C}}$	CONCENTRIC TO DATUM C WITHIN 0.0005 DIAMETER
$\boxed{\text{⊥}} \boxed{0.002} \boxed{\text{B}}$	PERPENDICULAR TO DATUM B WITHIN 0.002	$\boxed{\text{≡}} \boxed{0.010} \boxed{\text{A}}$	SYMMETRICAL WITH DATUM A WITHIN 0.010
$\boxed{\text{//}} \boxed{0.002} \boxed{\text{A}}$	PARALLEL TO DATUM A WITHIN 0.002	$\boxed{\text{∠}} \boxed{0.005} \boxed{\text{A}}$	ANGULAR TOLERANCE 0.005 WITH DATUM A
$\boxed{\text{○}} \boxed{0.002}$	ROUND WITHIN 0.002	$\boxed{\text{⊕}} \boxed{\text{∅}} \boxed{0.002} \boxed{\text{Ⓢ}} \boxed{\text{B}}$	LOCATED AT TRUE POSITION WITHIN 0.002 DIA RELATIVE TO DATUM B, REGARDLESS OF FEATURE SIZE
$\boxed{\text{⊘}} \boxed{0.010}$	CYLINDRICAL SURFACE MUST LIE BETWEEN TWO CONCENTRIC CYLINDERS, ONE OF WHICH HAS A RADIUS 0.010 INCH GREATER THAN THE OTHER	$\boxed{\text{⊥}} \boxed{\text{∅}} \boxed{0.010} \boxed{\text{Ⓜ}} \boxed{\text{A}}$	AXIS IS TOTALLY WITHIN A CYLINDER OF 0.010 INCH DIAMETER, PERPENDICULAR TO DATUM A, AND EXTENDING 0.510 INCH ABOVE DATUM A, MAXIMUM MATERIAL CONDITION
$\boxed{\text{⌒}} \boxed{0.006} \boxed{\text{A}}$	EACH LINE ELEMENT OF THE SURFACE AT ANY CROSS SECTION MUST LIE BETWEEN TWO PROFILE BOUNDARIES 0.006 INCH APART RELATIVE TO DATUM A	$\boxed{0.510} \boxed{\text{Ⓟ}}$	THEORETICALLY EXACT DIMENSION IS 2.000
$\boxed{\text{⌒}} \boxed{0.020} \boxed{\text{A}}$	SURFACES MUST LIE WITHIN PARALLEL BOUNDARIES 0.020 INCH APART AND EQUALLY DISPOSED ABOUT TRUE PROFILE		OR 2.000 BSC

True Position Dimensioning Symbols
Figure 601

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REPAIR - GENERAL

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REFINISH OF OTHER PARTS - REPAIR 1-1

1. General

- A. This procedure has the data necessary to refinish the parts which are not given in the specified repairs.
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to IPL Figure 1 for item numbers.

2. Refinish of Other Parts

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00033	Coating - Exterior Protective Enamel, Flexibility Use	BMS10-60, Type II
C00175	Primer - Urethane Compatible, Corrosion Resistant (Less Than 1% Aromatic Amines)	BMS10-79, Type III

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

C. General

- (1) Instructions for the repair of the parts listed in REPAIR 1-1, Table 601 is for repair of the initial finish.

D. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For finishing materials, refer to SOPM 20-60-02.

- (1) Refer to REPAIR 1-1, Table 601 for refinish details.

Table 601: Refinish Details

IPL FIG. & ITEM	MATERIAL	FINISH
IPL Fig. 1		
Seal retainer (60)	Aluminum alloy	Chemical treat (F-17.07). Apply primer, C00175 (F-19.47). Apply enamel coating, C00033 (F-19.39-707).
Clip (110, 115)	Aluminum alloy	Anodize (F-17.31). Apply primer, C00175 (F-19.47).
Bond assembly (120, 125)		Apply primer, C00175 (F-19.47).

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REPAIR 1-1

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END RIB - REPAIR 2-1

113A4112-17, -18, -19, -20

1. General

- A. This procedure has the data necessary to repair and refinish the end rib (165, 170, 175, 180).
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 for item numbers.
- E. General repair details:
 - (1) Material: Aluminum alloy

2. End Rib Repair

A. References

Reference	Title
SOPM 20-20-02	PENETRANT METHODS OF INSPECTION
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

B. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For finishing materials, refer to SOPM 20-60-02.

- (1) Ream the holes for the bushings (10, 15) to remove defects, cracks and/or corrosion up to the limit shown in REPAIR 2-1, Figure 602 and REPAIR 3-2, Figure 601.
- (2) Break all the sharp edges.
- (3) Do a penetrant check (SOPM 20-20-02) on the areas you reamed.
- (4) Make the oversize bushings as shown in REPAIR 2-1, Figure 603 and the steps shown below.
 - (a) Bushing (10) material: 15-5PH or 17-4PH CRES, 180-200 ksi.
 - (b) Bushing (15) material: Aluminum-nickel-bronze.
 - (c) Break all sharp edges.
 - (d) Bushing (10, 15) finish: Cadmium or zinc-nickel plate.
- (5) Install the oversize bushings as shown in REPAIR 3-1, Paragraph 2.C.(2).

3. End Rib Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00195	Adhesive - Corrosion Inhibiting Coating, Adhesive Primer	BMS5-89

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REPAIR 2-1

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B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure (REPAIR 2-1, Figure 601)

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For finishing materials, refer to SOPM 20-60-02.

- (1) Anodize (F-20.31).
- (2) Apply adhesive, A00195 (F-20.26) on the end rib (165, 170, 175, 180) but not in the bushing bores.

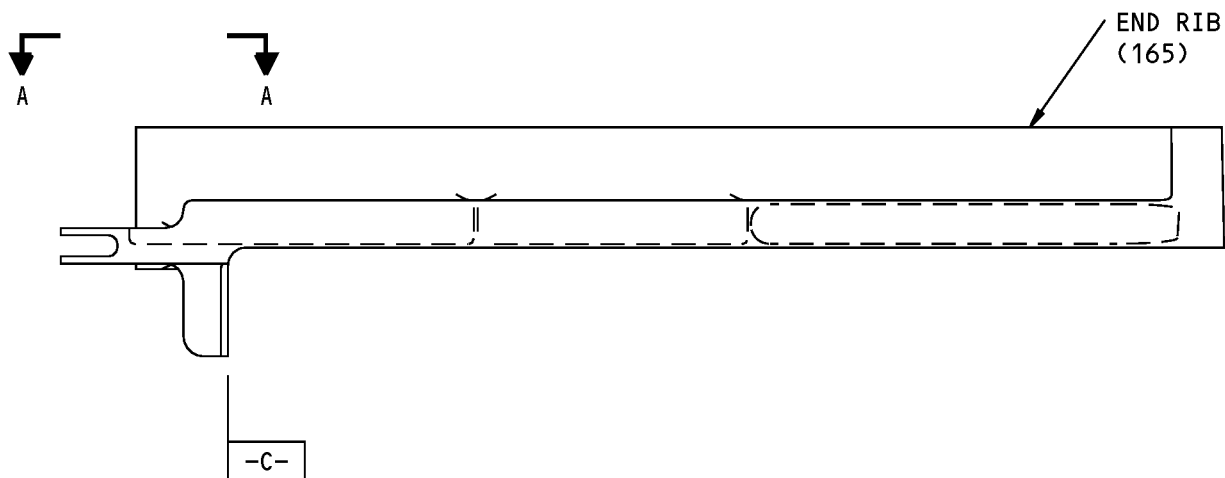
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REPAIR 2-1

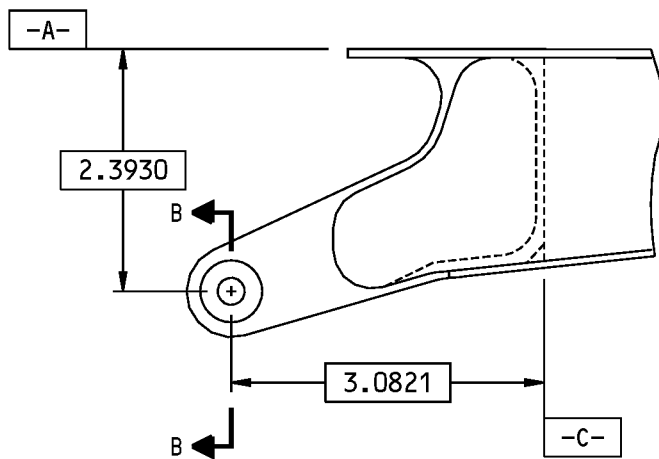
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113A4112-17 SHOWN
113A4112-18 OPPOSITE



A-A

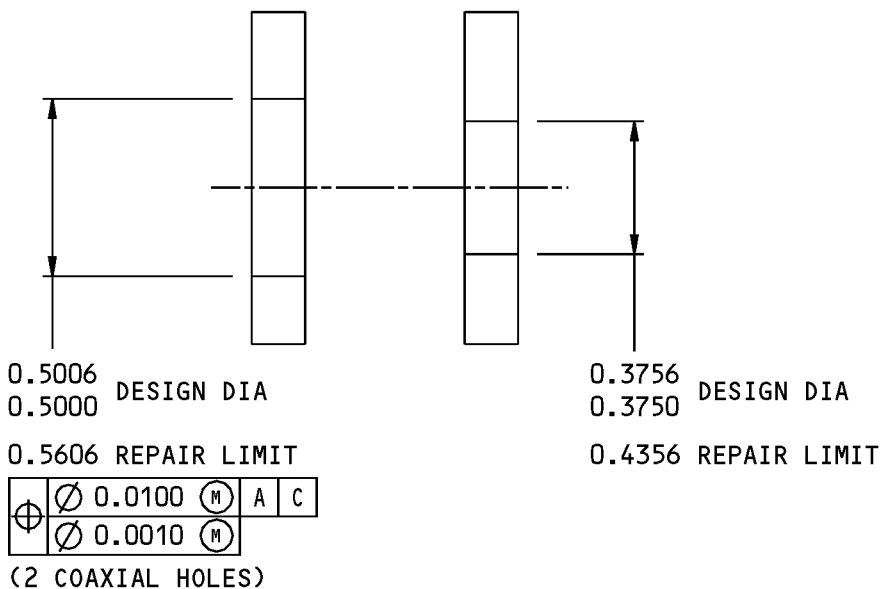
113A4112-17,-18 End Rib Repair
Figure 601 (Sheet 1 of 2)

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REPAIR 2-1
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B-B

63/√ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGE

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

113A4112-17,-18 End Rib Repair
Figure 601 (Sheet 2 of 2)

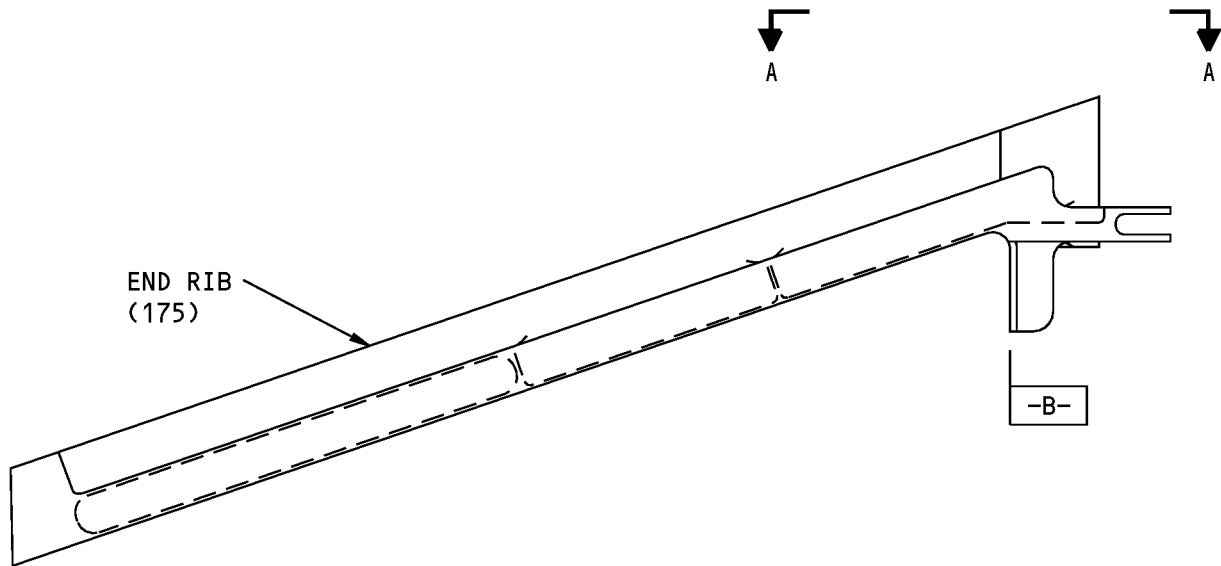
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REPAIR 2-1

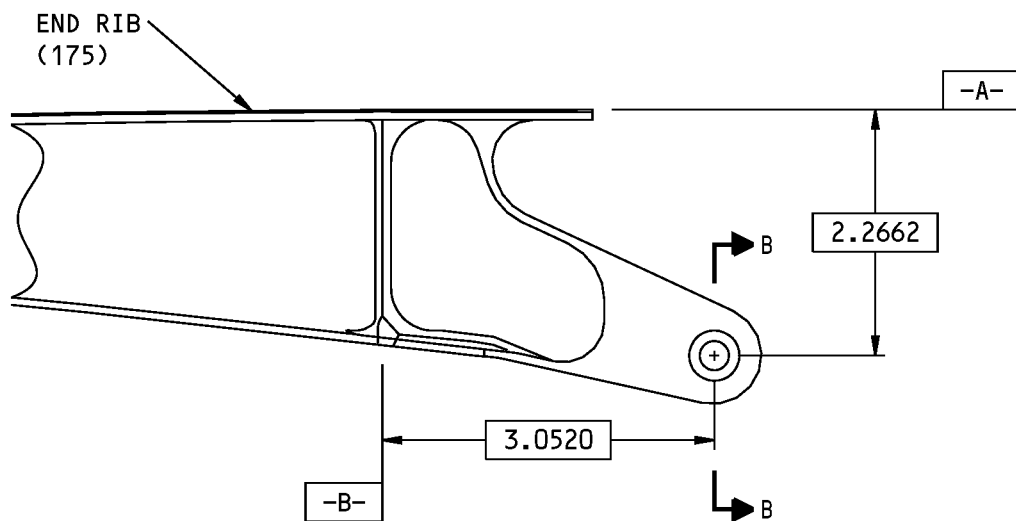
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COMPONENT MAINTENANCE MANUAL



113A4112-19 SHOWN
113A4112-20 OPPOSITE



A-A

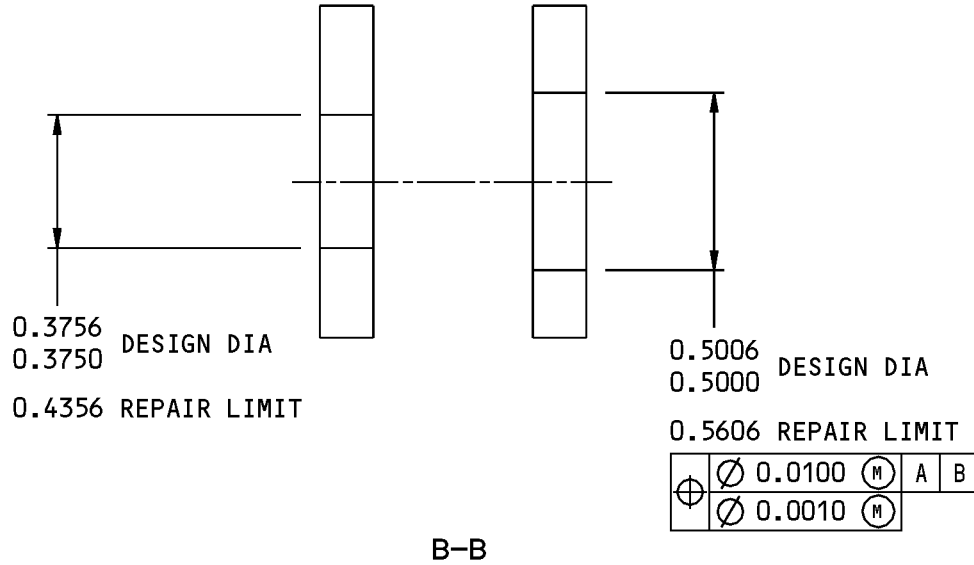
113A4112-19,-20 End Rib Repair
Figure 602 (Sheet 1 of 2)

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REPAIR 2-1
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COMPONENT MAINTENANCE MANUAL



125/√ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGE

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

113A4112-19,-20 End Rib Repair
Figure 602 (Sheet 2 of 2)

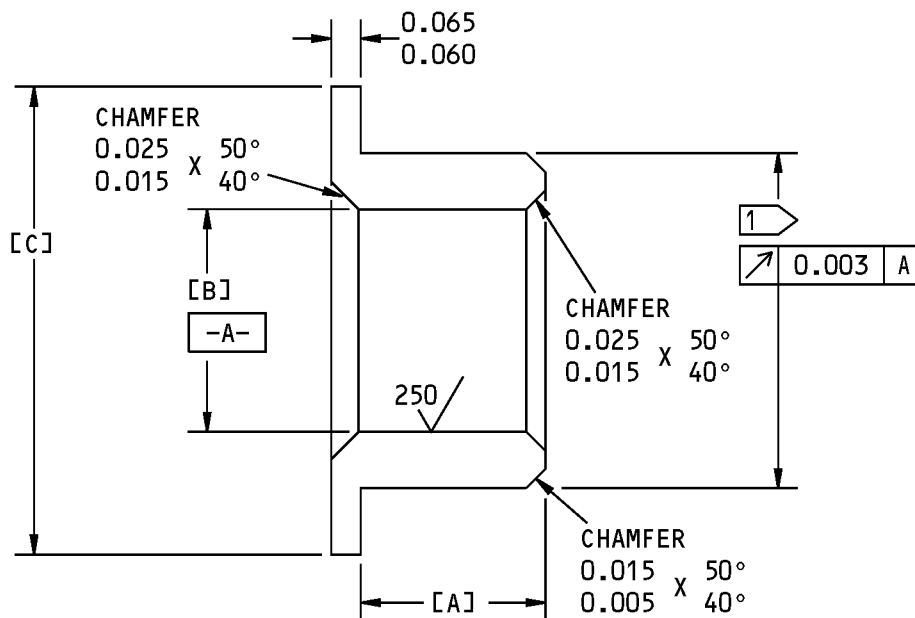
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REPAIR 2-1

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BUSHING TO BE REPLACED (IPL FIG. 1)	[A]	[B]	[C]	INTERFERENCE
10	0.140 0.135	0.2505 0.2500	0.540 0.530	0.0015 0.0003
15	0.140 0.135	0.3756 0.3750	0.630 0.620	0.0016 0.0004

1 THE OUTSIDE DIAMETER OF THE BUSHING AFTER PLATING IS EQUAL TO THE INSIDE DIAMETER OF THE LUG HOLE PLUS THE INTERFERENCE.

63/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGES

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

Oversize Bushing Details
Figure 603

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REPAIR 2-1

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NO. 5 AND 8 FLIGHT SPOILER ASSEMBLY - REPAIR 3-1

113A4500-1, -2, -3, -4

1. General

- A. This procedure has the data necessary to repair and refinish the No. 5 and 8 flight spoiler assembly (1A, 5).
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 for item numbers.

2. Bushing (10, 15) Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

- B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

- C. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bushings (10, 15) from the end rib (165, 170, 175, 180).
- (2) Install the new bushings (10, 15) on the end rib (165, 170, 175, 180) with sealant, A00247. Use the shrink fit method (SOPM 20-50-03).
- (3) Ream the inside diameter of the bushings (10, 15) to the dimensions shown in REPAIR 3-1, Figure 601.
- (4) Break all sharp edges.

3. Bearing (20) Replacement

- A. Consumable Materials

NOTE: Equivalent substitutes may be used.

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Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bearing (20) from the center fitting (185, 190).
- (2) Install the new bearing (20) on the center fitting (185, 190) with sealant, A00247 and roller swage (SOPM 20-50-03).

4. Bushing (70) Replacement

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95

B. References

Reference	Title
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-50-03	BEARING AND BUSHING REPLACEMENT
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure

NOTE: For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Remove the bushing (70) from the center fitting (185, 190).
- (2) Install the new bushing (70) on the center fitting (185, 190) with sealant, A00247. Use the shrink-fit method (SOPM 20-50-03).

5. Aluminum Foil Marker (205) Replacement

A. References

Reference	Title
SOPM 20-50-05	APPLICATION OF ALUMINUM FOIL AND OTHER MARKERS

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REPAIR 3-1
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B. Procedure

- (1) Remove the damaged aluminum foil marker (205) from the panel bond assembly (120).
- (2) Clean the surface.
- (3) Install the new aluminum foil marker (205) on the panel bond assembly (120). See REPAIR 3-1, Figure 602 and SOPM 20-50-05.

6. Seal (75) Replacement

A. References

Reference	Title
SOPM 20-50-12	APPLICATION OF ADHESIVES

B. Procedure

- (1) Remove the damaged seal (75) from the panel bond assembly (120).
- (2) Clean the surface.
- (3) Install the new seal (75) on the panel bond assembly (120). Bond as shown in SOPM 20-50-12.

7. No. 5 and 8 Flight Spoiler Assembly Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
C00033	Coating - Exterior Protective Enamel, Flexibility Use	BMS10-60, Type II

B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-02	FINISHING MATERIALS

C. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For finishing materials, refer to SOPM 20-60-02.

- (1) Apply enamel coating, C00033(F-19.39-707). No finish on bearings (20), bushings (10, 15, 70), seals (65, 70), seal retainers (60), and rubstrip (130).

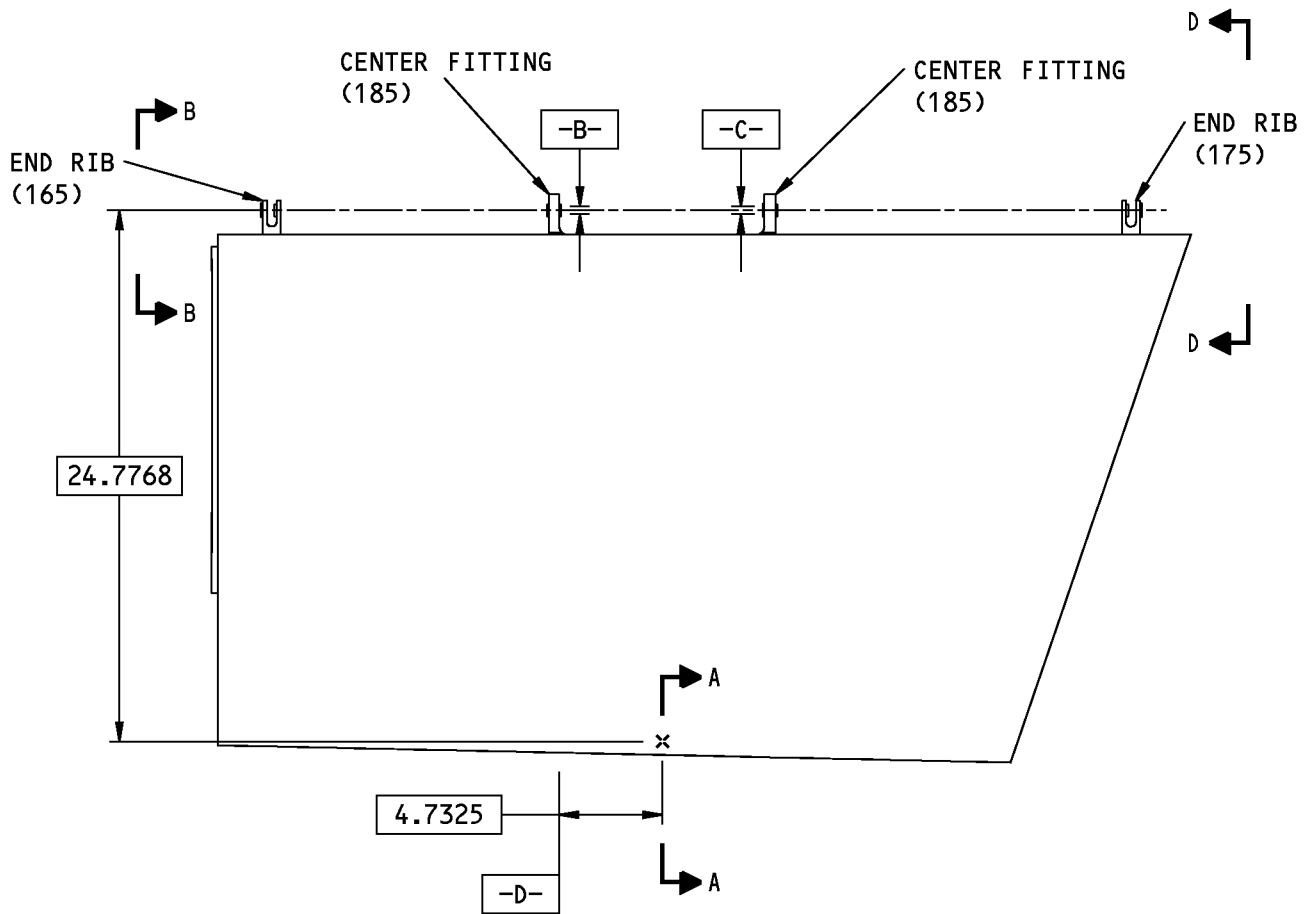
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REPAIR 3-1

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113A4500-1,-3 SHOWN
 113A4500-2,-4 OPPOSITE

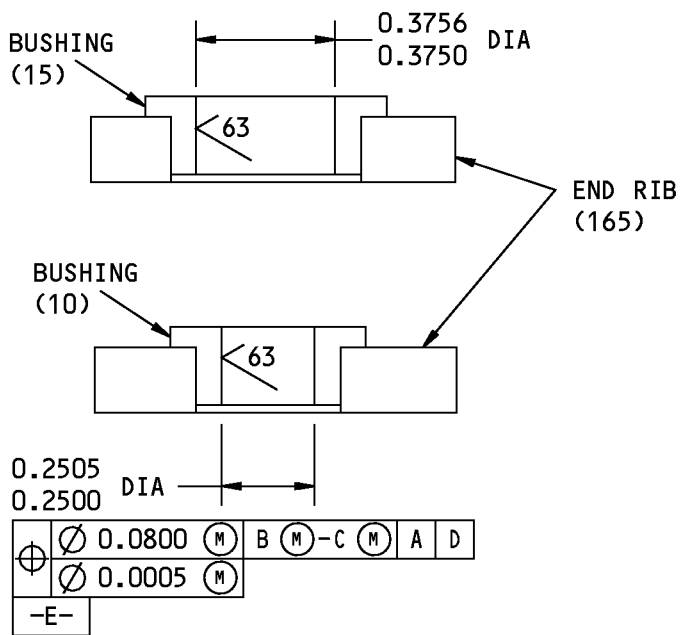
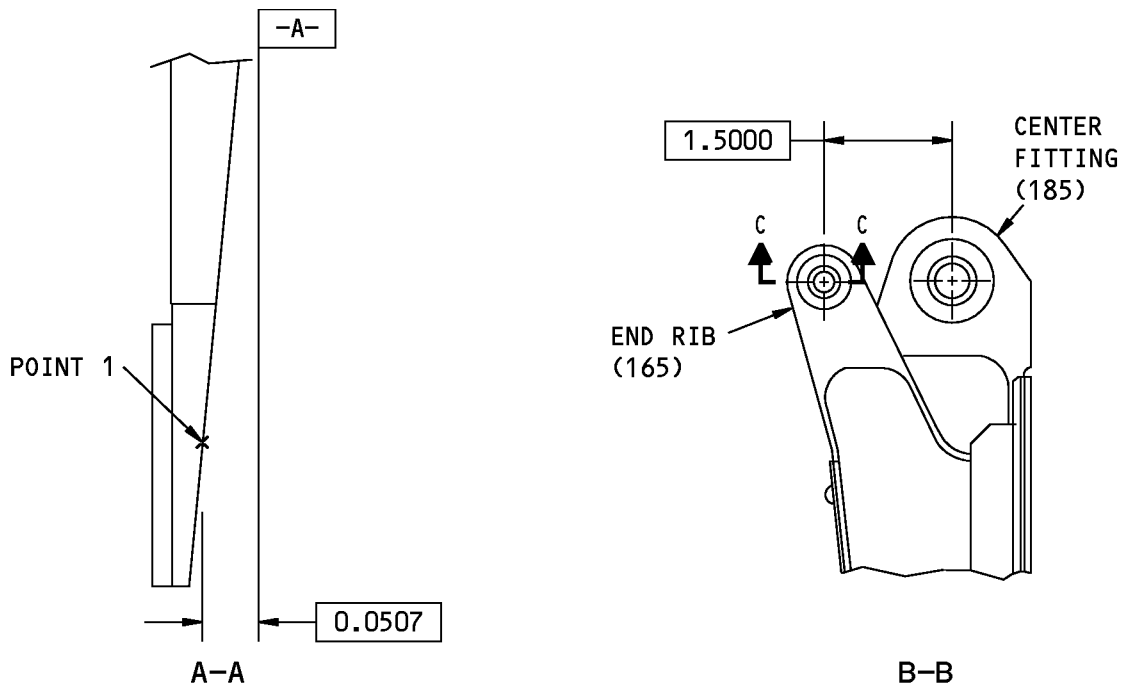
G24307 S00041007835_V2

113A4500-1,-2,-3,-4 No. 5 and 8 Flight Spoiler Assembly Repair
 Figure 601 (Sheet 1 of 3)

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REPAIR 3-1
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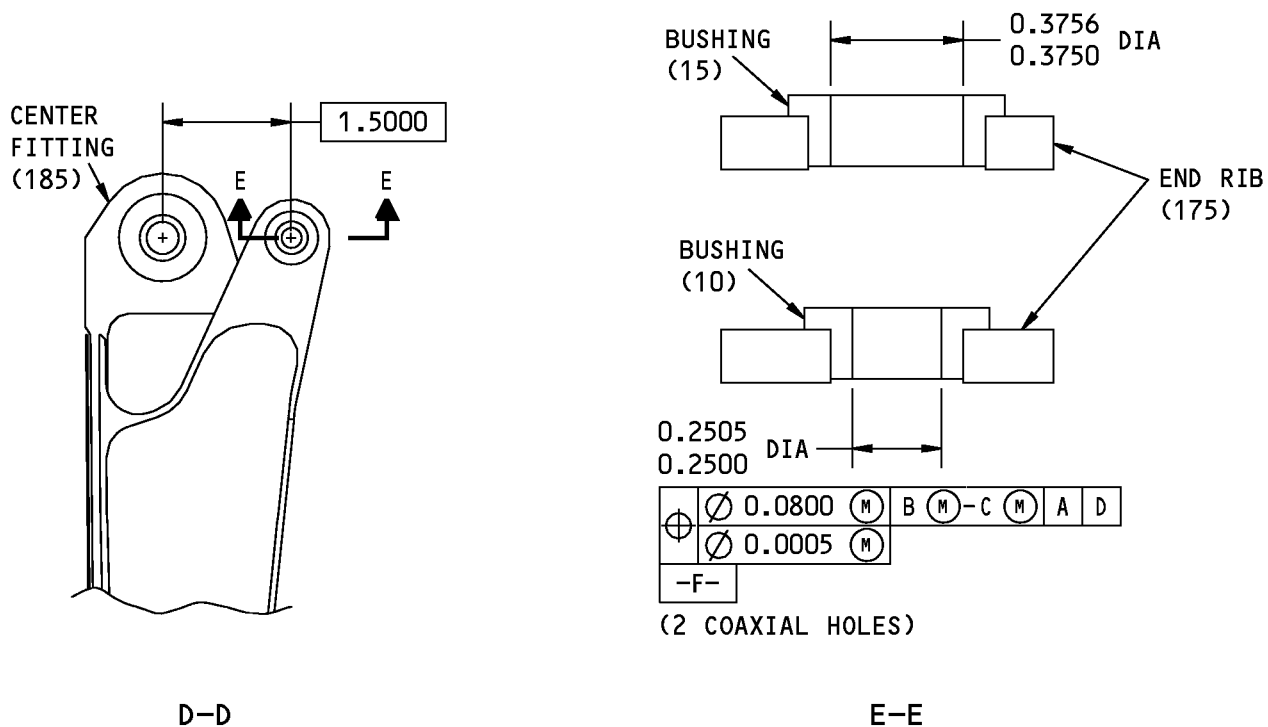
G24318 S00041007836_V2

113A4500-1,-2,-3,-4 No. 5 and 8 Flight Spoiler Assembly Repair
Figure 601 (Sheet 2 of 3)

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REPAIR 3-1
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63/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGES

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

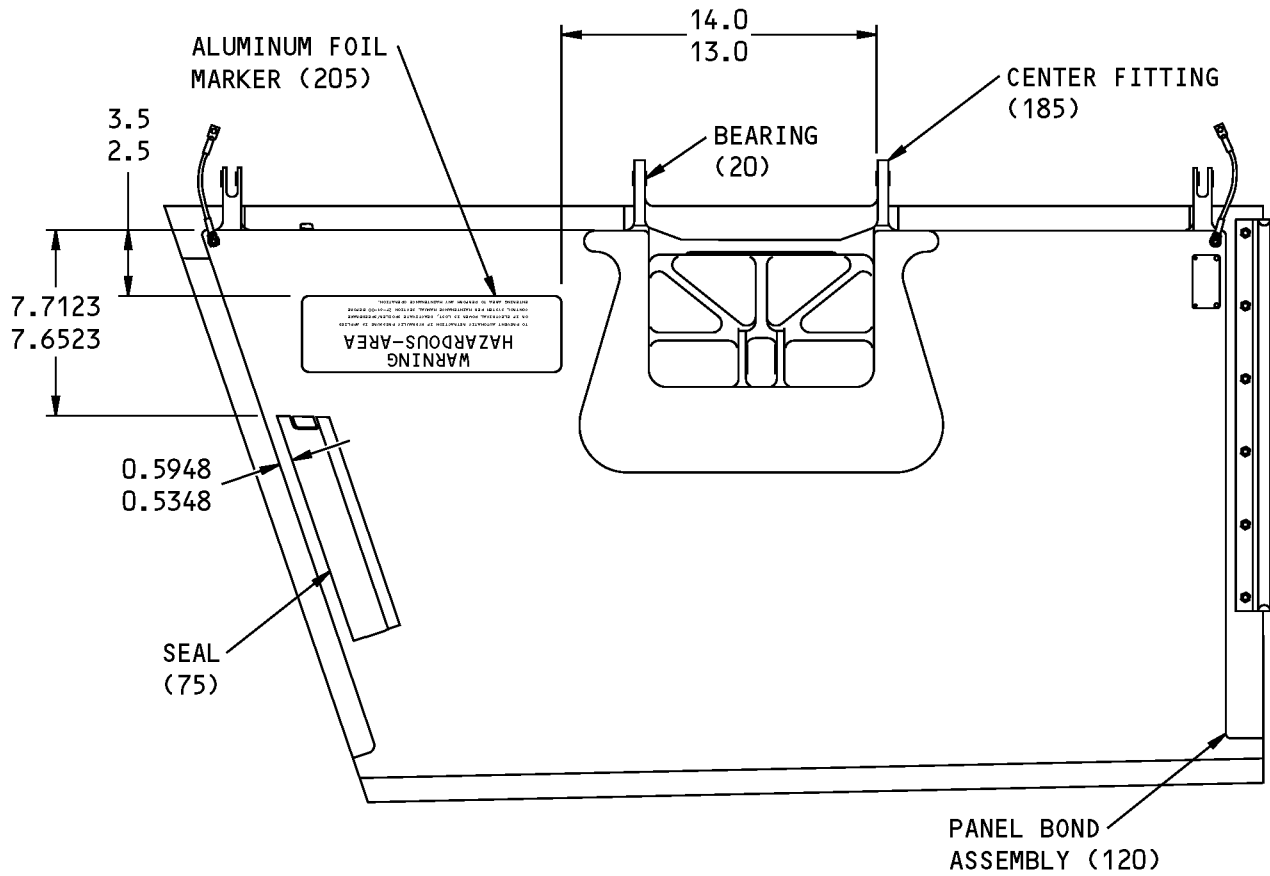
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113A4500-1,-2,-3,-4 No. 5 and 8 Flight Spoiler Assembly Repair
Figure 601 (Sheet 3 of 3)

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REPAIR 3-1
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ITEM NUMBERS REFER TO IPL FIG. 1
ALL DIMENSIONS ARE IN INCHES

Miscellaneous Parts Replacement
Figure 602

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REPAIR 3-1
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FRAME ASSEMBLY - REPAIR 3-2

113A4510-3, -4

1. General

- A. This procedure has the data necessary to repair and refinish the frame assembly (135, 140).
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 for item numbers.

2. End Rib Repair

- A. Procedure
 - (1) See REPAIR 3-2, Figure 601 and REPAIR 3-1 for repair and refinish of the end rib (165, 170, 175, 180).

3. Center Fitting Repair

- A. Procedure
 - (1) See REPAIR 3-2, Figure 601 and REPAIR 4-1 for repair and refinish of the center fitting (185, 190).

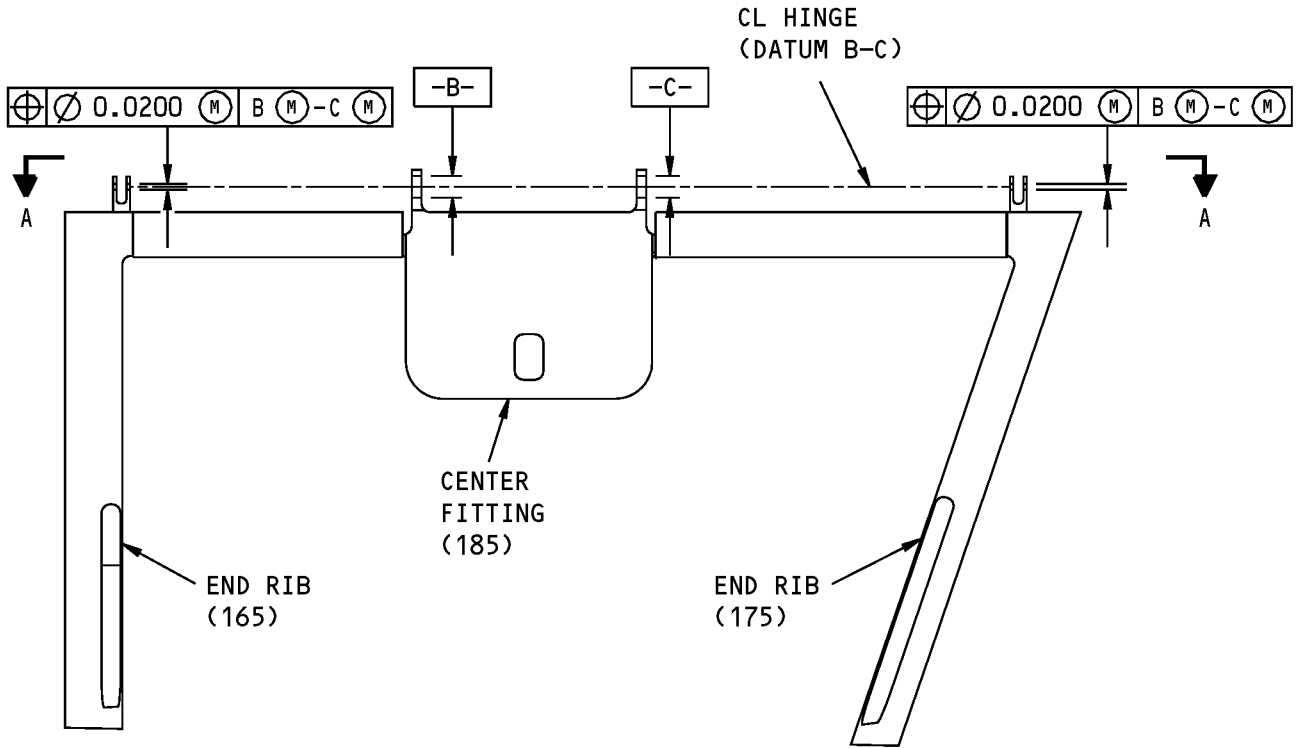
57-56-56

REPAIR 3-2

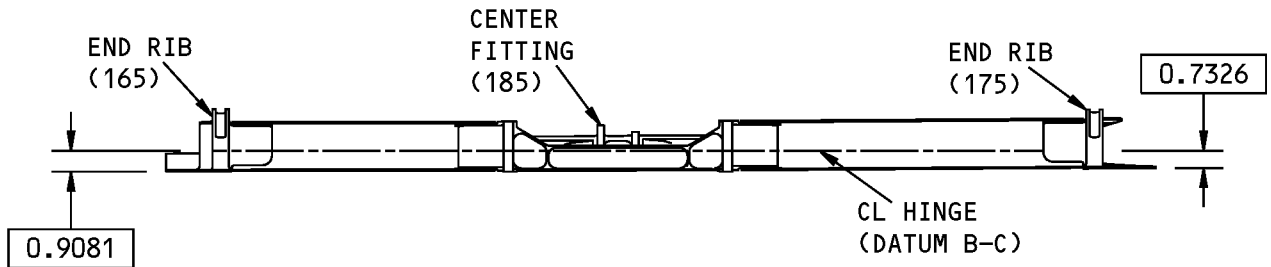
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113A4510-3 SHOWN
113A4510-4 OPPOSITE



A-A

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGES

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

113A4510-3,-4 Frame Assembly Repair
Figure 601

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REPAIR 3-2

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CENTER FITTING - REPAIR 4-1

113A4520-1, -2

1. General

- A. This procedure has the data necessary to repair and refinish the center fitting (185, 190).
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to REPAIR-GENERAL, Figure 601 for the Standard True Position Dimensioning Symbols shown in the repair.
- D. Refer to IPL Figure 1 for item numbers.
- E. General repair details:
 - (1) Material: Aluminum alloy

2. Center Fitting (185, 190) Repair

A. References

Reference	Title
SOPM 20-20-02	PENETRANT METHODS OF INSPECTION
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES

B. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01.

- (1) Ream the holes for the bushings (70) to remove defects, cracks and/or corrosion up to the limit shown in REPAIR 4-1, Figure 601, REPAIR 4-1, Figure 602 and REPAIR 3-1, Figure 601.
- (2) Break all sharp edges.
- (3) Do a penetrant check (SOPM 20-20-02) on the areas you reamed.
- (4) Make the oversize bushings as shown in REPAIR 4-1, Figure 602 and the steps shown below.
 - (a) Bushing (70) material: Aluminum-nickel-bronze.
 - (b) Break all sharp edges.
 - (c) Bushing (70) finish: Cadmium plate.
- (5) Install the oversize bushings as shown in REPAIR 3-1, Paragraph 2.C.(2).

3. Center Fitting (185, 190) Refinish

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00195	Adhesive - Corrosion Inhibiting Coating, Adhesive Primer	BMS5-89

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REPAIR 4-1
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B. References

Reference	Title
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For decoding table for Boeing finish codes, refer to SOPM 20-41-01. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Anodize (F-20.31).
- (2) Apply adhesive, A00195 (F-20.26) on the center fitting (185, 190) but not in the bushing bores.

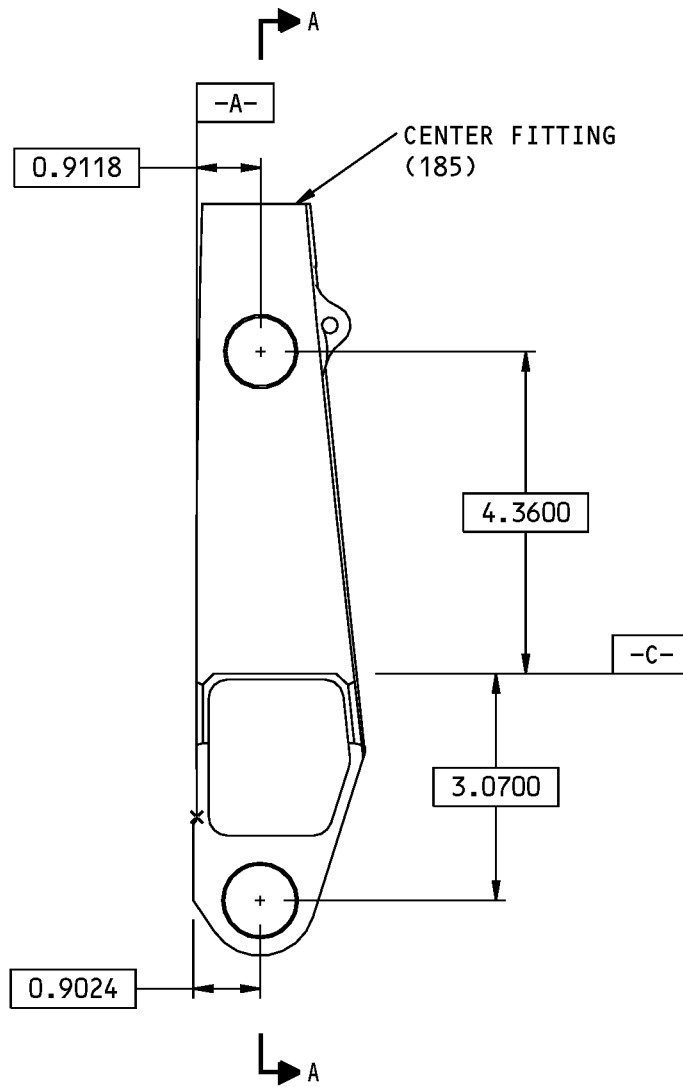
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REPAIR 4-1

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113A4520-1 SHOWN
113A4520-2 OPPOSITE

113A4520-1,-2 Center Fitting Repair
Figure 601 (Sheet 1 of 3)

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REPAIR 4-1
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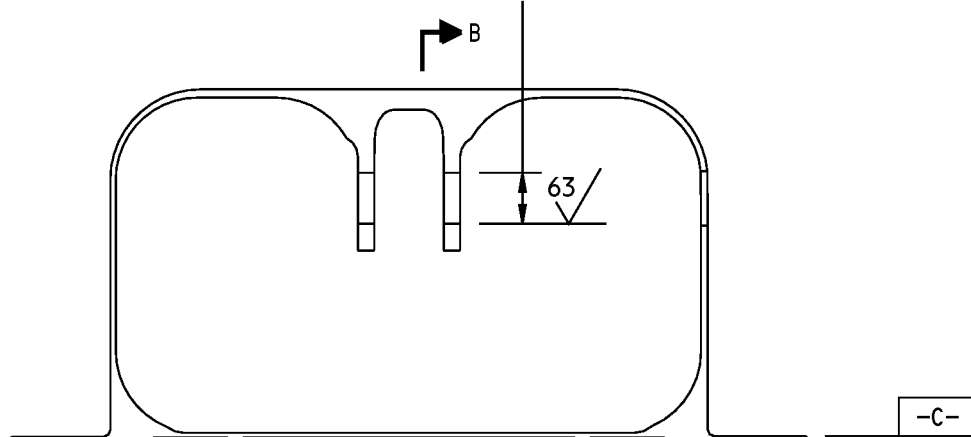
COMPONENT MAINTENANCE MANUAL

0.9379
0.9370 DESIGN DIA

0.9979 REPAIR LIMIT

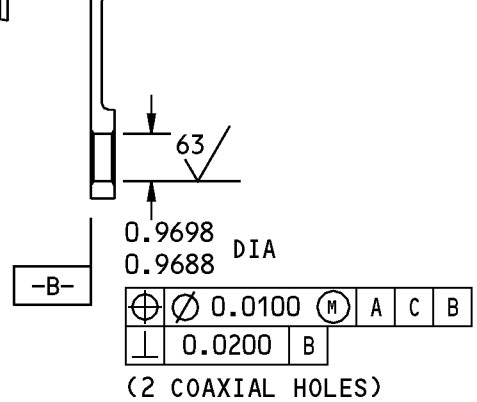
\varnothing 0.0100	(M)	A	C
\oplus	0.0015	(M)	

(2 COAXIAL HOLES)



CHAMFER
0.0300
0.0250 X 45°
(4 LOCATIONS)

-C-



0.9698
0.9688 DIA

\oplus	\varnothing 0.0100	(M)	A	C	B
\perp	0.0200	B			

(2 COAXIAL HOLES)

A-A

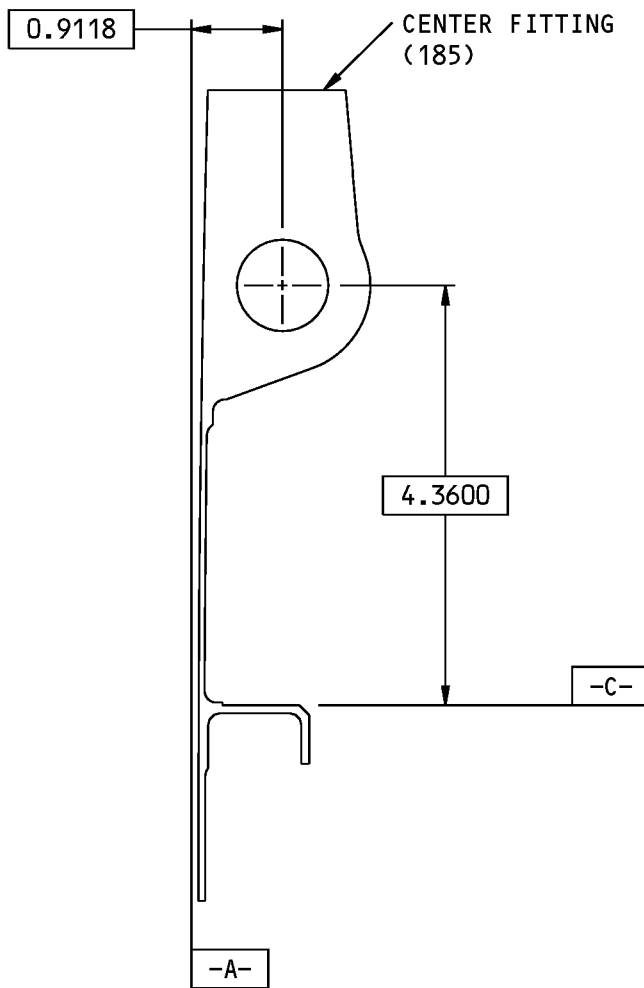
113A4520-1,-2 Center Fitting Repair
Figure 601 (Sheet 2 of 3)

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REPAIR 4-1
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B-B

125/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGE

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

113A4520-1,-2 Center Fitting Repair
Figure 601 (Sheet 3 of 3)

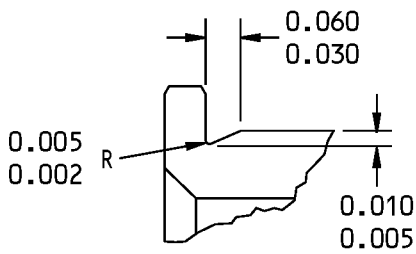
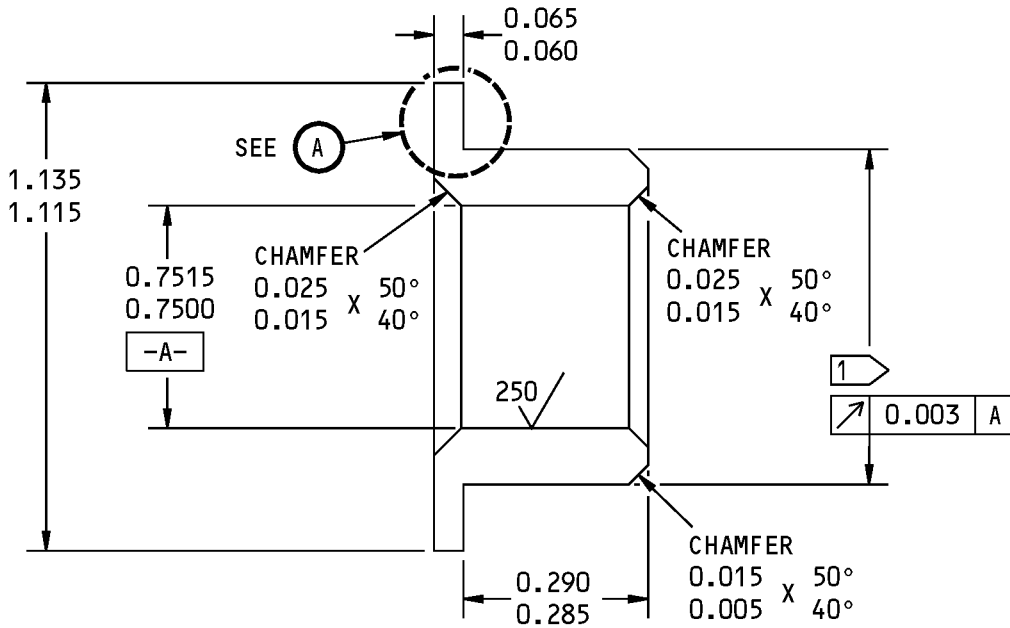
57-56-56

REPAIR 4-1

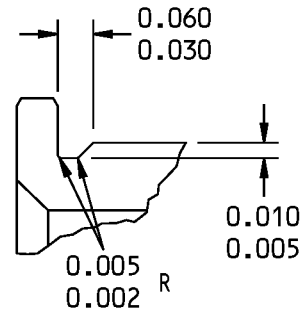
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OR



(A)

1 THE OUTSIDE DIAMETER OF THE BUSHING AFTER PLATING IS EQUAL TO THE INSIDE DIAMETER OF THE LUG HOLE PLUS THE INTERFERENCE OF 0.0004-0.0023.

63/ ALL MACHINED SURFACES UNLESS SHOWN DIFFERENTLY

BREAK ALL SHARP EDGES

ITEM NUMBERS REFER TO IPL FIG. 1

ALL DIMENSIONS ARE IN INCHES

Oversize Bushing Details
Figure 602

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REPAIR 4-1

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ASSEMBLY

1. General

- A. This procedure has the data necessary to assemble the No. 5 and 8 flight spoiler assembly.
- B. Refer to the Standard Overhaul Practices Manual (SOPM) for the SOPM subjects identified in this procedure.
- C. Refer to IPL Figure 1 for item numbers.

2. Assembly

A. Consumable Materials

NOTE: Equivalent substitutes may be used.

Reference	Description	Specification
A00247	Sealant - Pressure And Environmental - Chromate Type	BMS 5-95
C00259	Primer - Chemical And Solvent Resistant Finish, Epoxy Resin	BMS10-11, Type I
C00528	Compound - Corrosion Preventive, Petroleum Hot Application (Soft Film)	MIL-C-11796, Class III
C00862	Coating - Chemical Conversion - Alodine 600	

B. References

Reference	Title
SOPM 20-11-03	REPAIR OF ELECTRICAL TERMINATIONS AND ELECTRICAL BONDING AREAS
SOPM 20-30-02	STRIPPING OF PROTECTIVE FINISHES
SOPM 20-41-01	DECODING TABLE FOR BOEING FINISH CODES
SOPM 20-41-02	APPLICATION OF CHEMICAL AND SOLVENT RESISTANT FINISHES
SOPM 20-43-03	CHEMICAL CONVERSION COATINGS FOR ALUMINUM
SOPM 20-50-01	BOLT AND NUT INSTALLATION
SOPM 20-60-02	FINISHING MATERIALS
SOPM 20-60-03	LUBRICANTS
SOPM 20-60-04	MISCELLANEOUS MATERIALS

C. Procedure

NOTE: For stripping of protective finishes, refer to SOPM 20-30-02. For the decoding table for Boeing finish codes, refer to SOPM 20-41-01. For bolt and nut installation, refer to SOPM 20-50-01. For finishing materials, refer to SOPM 20-60-02. For lubricants, refer to SOPM 20-60-03. For miscellaneous materials, refer to SOPM 20-60-04.

- (1) Use standard industry procedures and the steps shown below to assemble this component.
- (2) Install the jumper assemblies (40) on the panel bond assembly (120, 125) (SOPM 20-11-03).
 - (a) Remove the finish, if necessary, in areas where you will install the jumper assemblies (40).

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ASSEMBLY

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- (b) Apply Alodine 600 coating, C00862 (SOPM 20-43-03) by hand, by brush, swab, or spray application (F-17.28) in areas you just removed the finish.
 - (c) Install the jumper assemblies (40) on the panel bond assembly (120, 125) with bolts (25), washers (30), and nuts (35) (SOPM 20-11-03). Use three washers for installation, one washer (30) under the bolt head, and one washer (30) on each side of the jumper assembly (40).
 - (d) Fillet seal the ends of the jumper assemblies (40) with sealant, A00247.
- (3) Install the seals (65) on the panel bond assembly (120, 125).
- (a) In areas around the holes for bolts (45), apply primer, C00259 (SOPM 20-41-02) to all areas of the holes and countersink, counterbore, or other recess.
NOTE: Do the next step immediately after you complete this step.
 - (b) In areas around the holes for bolts (45), apply compound, C00528 to all areas of the holes and countersink, counterbore, or other recess.
NOTE: Do the next step immediately after you complete this step.
 - (c) Install the seal (65) and seal retainer (60) on the panel bond assembly (120, 125) with bolts (45), washers (50), and nuts (55).

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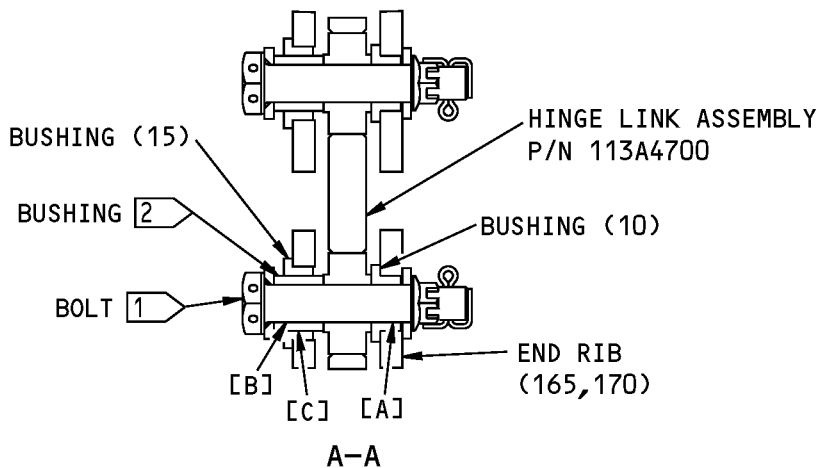
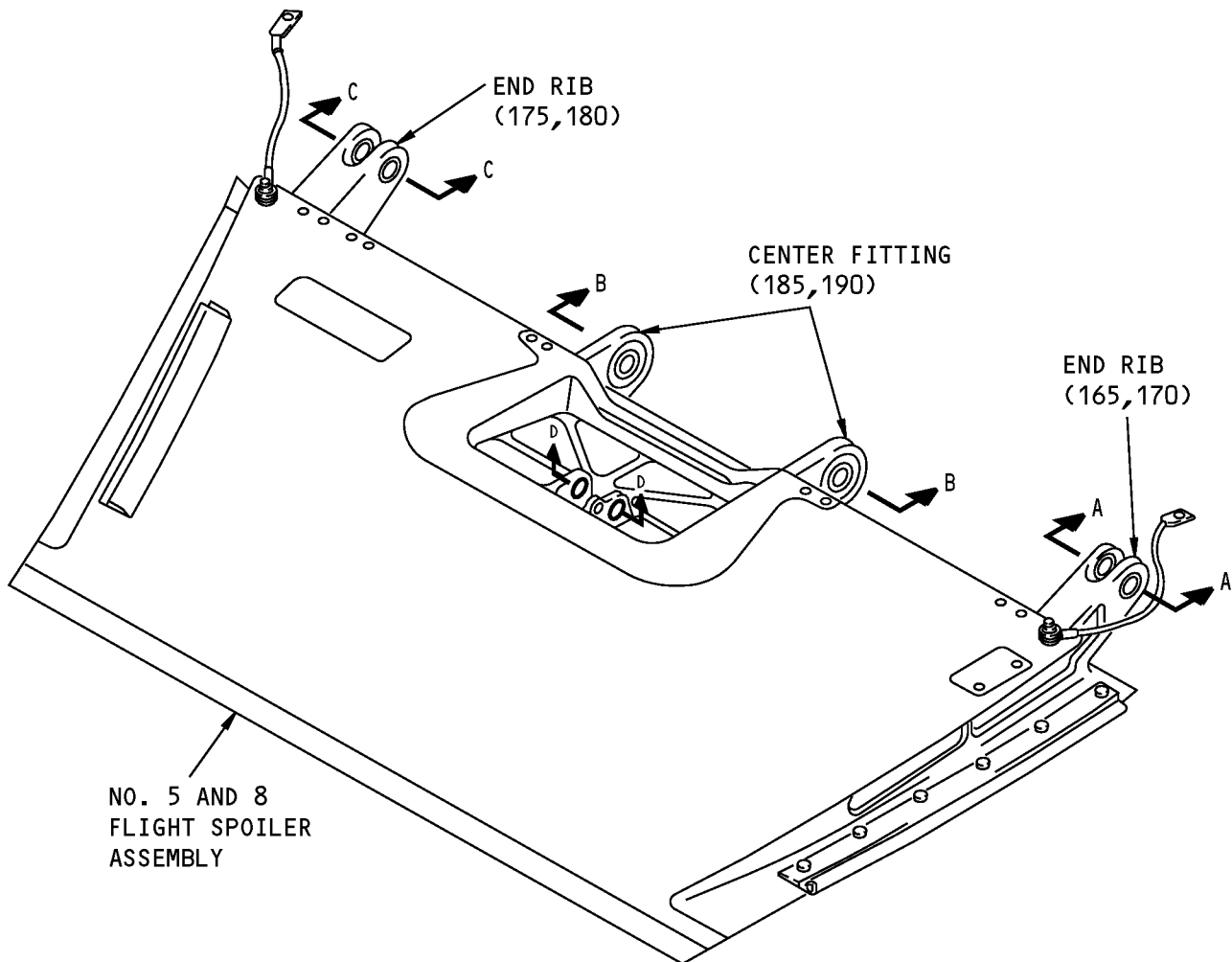
ASSEMBLY

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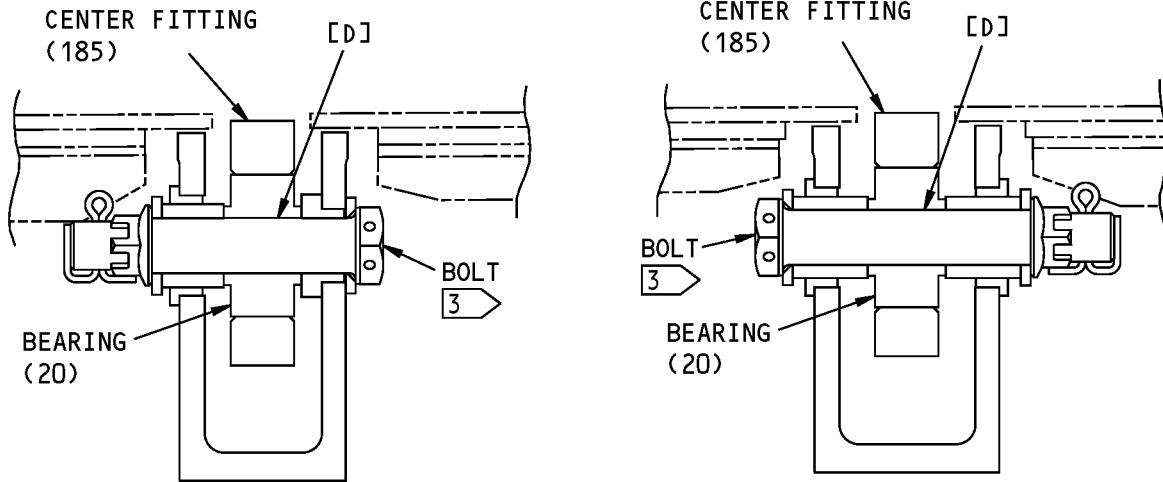
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FITS AND CLEARANCES

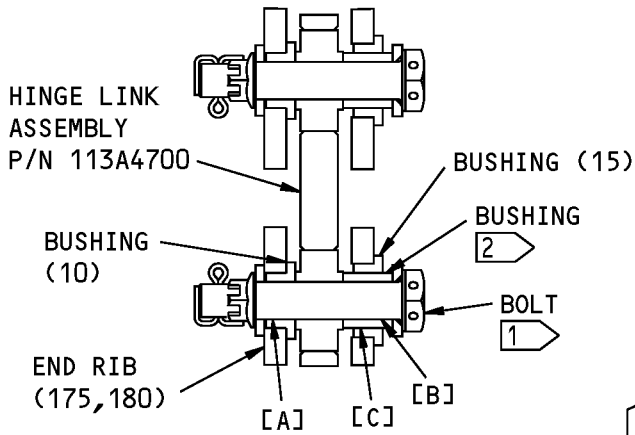


Fits and Clearances
Figure 801 (Sheet 1 of 3)

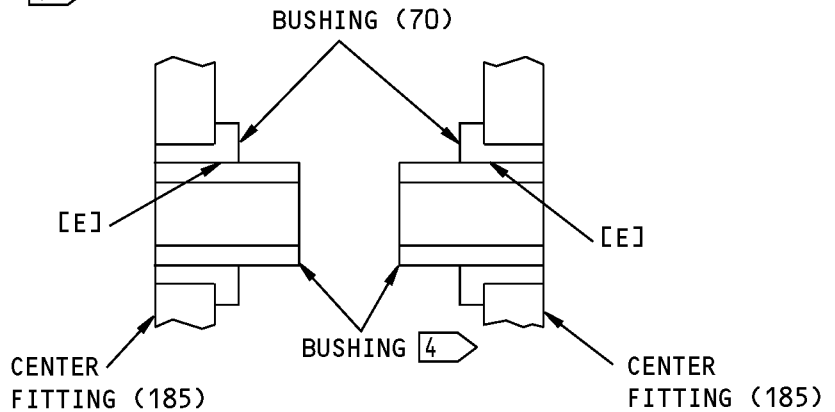
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B-B



C-C



D-D

Fits and Clearances
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




FITS AND CLEARANCES

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
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REF LETTER	REF IPL	DESIGN DIMENSION*				SERVICE WEAR LIMIT*		
	FIG. 1, MATING ITEM NO.	DIMENSION		ASSEMBLY CLEARANCE		DIMENSION		MAXIMUM CLEARANCE
		MIN	MAX	MIN	MAX	MIN	MAX	
[A]	ID BUSHING (10)	0.2500	0.2505	0.0005	0.0024	0.2476	0.2530	0.005
	OD BOLT 	0.2481	0.2495					
[B]	ID BUSHING 	0.2500	0.2505	0.0005	0.0024	0.2476	0.2530	0.005
	OD BOLT 	0.2481	0.2495					
[C]	ID BUSHING (15)	0.3750	0.3756	0.0005	0.0016	0.3700	0.3795	0.005
	OD BUSHING 	0.3740	0.3745					
[D]	ID BEARING (20)	0.3750	0.3755	0.0005	0.0024	0.3730	0.3780	0.005
	OD BOLT 	0.3731	0.3745					
[E]	ID BUSHING (25)	0.7500	0.7515	0.0010	0.0030	0.7465	0.7775	0.005
	OD BUSHING 	0.7485	0.7490					

* ALL DIMENSIONS ARE IN INCHES

-  INSTALLATION BOLT, P/N
BACB30NM4DK16
-  INSTALLATION BUSHING, P/N
BACB28AK04-034
-  INSTALLATION BOLT, P/N
BACB30NM6DK22
INSTALLATION BOLT, P/N
BACB30NM6DK27
-  INSTALLATION BUSHING, P/N
BACB28BA1012063

Fits and Clearances
Figure 801 (Sheet 3 of 3)

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SPECIAL TOOLS, FIXTURES, AND EQUIPMENT

(NOT APPLICABLE)

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SPECIAL TOOLS, FIXTURES, AND EQUIPMENT

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ILLUSTRATED PARTS LIST

1. Introduction

- A. The Illustrated Parts List (IPL) contains an illustration and a list of component parts you can repair or replace. The Illustrated Parts Catalog (IPC) shows how to use the Boeing part number system.
- B. This shows how parts are related: The relation of each item to its next higher assembly (NHA) is shown in the NOMENCLATURE column. Use the indenture system that follows:

1	2	3	4	5	6	7
.	Assembly					
.	Attaching parts for assembly					
.	.	Detail parts for assembly				
.	.	Subassembly				
.	.	Attaching parts for subassembly				
.	.	.	Detail parts for subassembly			
.	.	.	Sub-subassembly			
.	.	.	Attaching parts for subassembly			
.	.	.	.	Details parts for sub-subassembly		
						Detail Installation Parts (Included only if installation parts may be sent to the shop as part of assembly)

- C. Each top assembly is given one use code letter (A, B, C, etc.) in the USAGE CODE column. All subsequent component parts in the list can have one or more of the use code letters to show effectivity to top assemblies. A component part without a use code applies to all top assemblies.
- D. An alphabetical letter is added after the item number for optional parts, parts changed by a Service Bulletin, configuration differences (except left-handed and right-handed parts), last engineering releases, and parts added between item numbers in a sequence. The alphabetical letter will not be shown on the illustration for equivalent parts of the same part number.
- E. Color-coded parts are identified with a single digit alpha following the dash number or with "SP" suffix. If the "SP" suffix is used, it represents consolidation of all color codes applicable for a given usage which are not separately listed. Orders for color-coded parts should include the registry number of the airplane for which the parts are ordered.
- F. If a part number is 15 characters long but will not fit in the part number column, the part number will be displayed with a "~" at the end of the line and will be continued on the next line. The "~" denotes that the part number continues on the next line.
- G. Parts changed by a Service Bulletin are shown by PRE SB XXXX and POST SB XXXX added to the NOMENCLATURE column.
- (1) When a new top assembly is added by a Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the top assembly level only. The configuration differences at the detail part level are shown by use code letters.
- (2) When the top assembly part number is not changed by the Service Bulletin, PRE SB XXXX and POST SB XXXX will be added at the detail level.
- H. Interchangeable Parts

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Optional (OPT)	The part is optional to and interchangeable with other parts that have the same item number.
Replaces, Replaced by and not interchangeable with (REPLACES, REPLACED BY AND NOT INTCHG/W)	The part replaces and is not interchangeable with the initial part.
Replaces, Replaced by (REPLACES, REPLACED BY)	The part replaces and is interchangeable with, or is an alternative to, the initial part.

VENDOR CODES

Code	Name
15653	ALCOA GLOBAL FASTENERS INC DIV KAYNAR PRODUCTS 800 S STATE COLLEGE BLVD FULLERTON, CALIFORNIA 92831-3001 FORMERLY VK6405 MICRODOT AEROSP LTD; FORMERLY KAYNAR TECH FORMERLY FAIRCHILD FASTENERS KAYNAR DIV
15860	NEW HAMPSHIRE BALL BEARINGS, INC ASTRO DIVISION 155 LEXINGTON AVENUE LACONIA, NEW HAMPSHIRE 03246-2937 FORMERLY ASTRO BEARING CORP, LOS ANGELES, CALIF.
1FF12	CIRCUIT SYSTEMS CO 2621 COLORADO CIR PO BOX 171322 ARLINGTON, TEXAS 76017
1GK47	R AND B ELECTRONICS INC 2374 NW DALLAS STREET GRAND PRAIRIE, TEXAS 75050
50632	KAMATICS CORP SUB OF KAMAN CORP 1335 BLUE HILLS ROAD BLOOMFIELD, CONNECTICUT 06002-1304
62554	SIMMONDS MECAERO FASTENERS INC 1734 SEQUOIA AVENUE ORANGE, CALIFORNIA 92668

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Code	Name
73134	<p>ROLLER BEARING COMPANY OF AMER DBA HEIM BEARINGS DIV 60 ROUND HILL RD FAIRFIELD, CONNECTICUT 06430-0000 FORMERLY INCOM INTL HEIM DIV; HEIM UNIVERSAL CORP INCOM; FORMERLY HEIM DIV INCOM INTL; IMO IND HEIM BEARINGS DIV</p>
81376	<p>SMITH ACQUISITION COMPANY 2240 BUENA VISTA BALDWIN PARK, CALIFORNIA 91706</p>
91812	<p>ESTERLINE MASON 13955 BALVOA ROAD SYLMAR, CALIFORNIA 91342 FORMERLY JANCO CORPORATION</p>
97613	<p>SARGENT CONTROLS & AEROSPACE/KAHR BEARING DIV 5675 W BURLINGAME RD TUCSON, ARIZONA 85743 FORMERLY AETNA STEEL PROD KAHR BEARING DIV V96579 FORMERLY SARGENT IND KAHR BEARING DIV, BURBANK, CALIFORNIA</p>
S0352	<p>NIPPON MINIATURE BEARING CO LTD TOKYO, JAPAN</p>

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NUMERICAL INDEX

PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
100220-7		1	40	2
113A4111-7		1	150	1
113A4111-9		1	155	1
113A4112-17		1	165	1
113A4112-18		1	170	1
113A4112-19		1	175	1
113A4112-20		1	180	1
113A4114-5		1	130	1
113A4150-11		1	65	1
113A4150-5		1	75	1
113A4150-6		1	80	1
113A4160-11		1	60A	1
113A4160-3		1	60	1
113A4170-1		1	110	1
113A4170-2		1	115	1
113A4500-1		1	1A	RF
113A4500-2		1	5	RF
113A4500-3		1	1B	RF
113A4500-4		1	5A	RF
113A4510-1		1	120	1
113A4510-2		1	125	1
113A4510-3		1	135	1
113A4510-4		1	140	1
113A4520-1		1	185	1
113A4520-2		1	190	1
940CW20-7		1	40	2
ADW06V301NC		1	20	2
BAC27NCT0217		1	205	1
BACB10FA06GC		1	20	2
BACB28AP04P014		1	10	2
BACB28AT06B014C		1	15	2
BACB28X12K029		1	70A	2
BACB28X12M029		1	70	2
BACB30VF08K4		1	45	6

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PART NUMBER	AIRLINE PART NUMBER	FIGURE	ITEM	UNITS PER ASSEMBLY
BACJ40AB20-7		1	40	2
BACN10JC08CD		1	55	6
BACN10YR3CD		1	35	2
BACR15BB3D		1	195	2
BACR15BB3D4C		1	195A	2
BACR15CE5D		1	90	2
BACR15CE6D		1	85	4
BACR15FT5D		1	105	2
BACR15FT5D5C		1	105A	2
BACR15FT6D		1	100	8
		1	145	16
BACR15GF5D5		1	90A	2
BACR15GF6D7		1	85A	4
BACS12GU3K7		1	25	2
BACS40R008B018F		1	160	AR
H52732-3CD		1	35	2
KSC152200BZ06GC		1	20	2
KWDB06-35		1	20	2
MS27253-1		1	200	1
NAS1149D0316J		1	30	6
NAS1149DN832J		1	50	6
PLH53CD		1	35	2
RBEJ40AB20-7		1	40	2
SWKRS06-350SC		1	20	2
WES06FAGC		1	20	2
WHTFA06VC		1	20	2

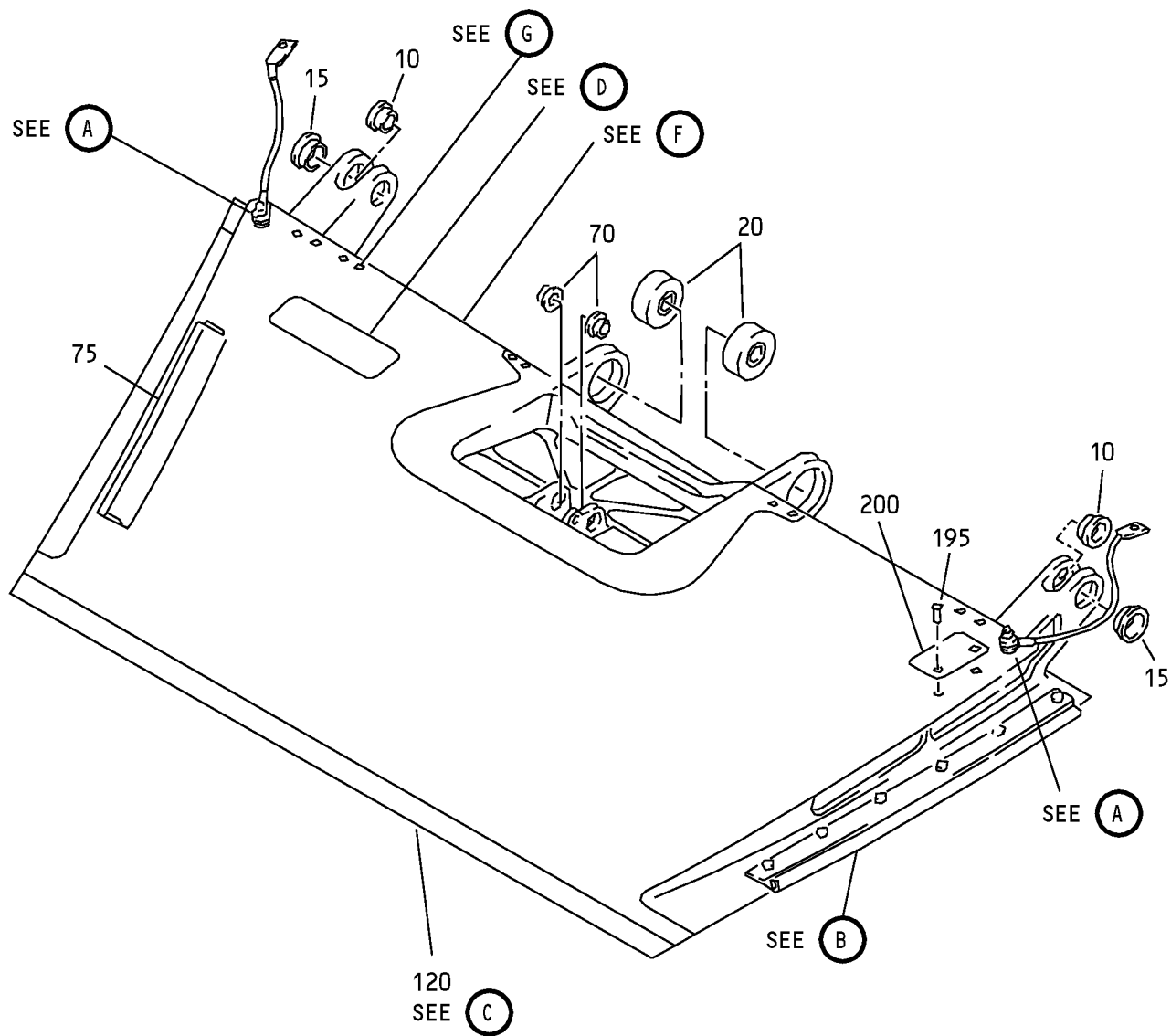
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No. 5 and 8 Flight Spoiler Assembly
IPL Figure 1 (Sheet 1 of 5)

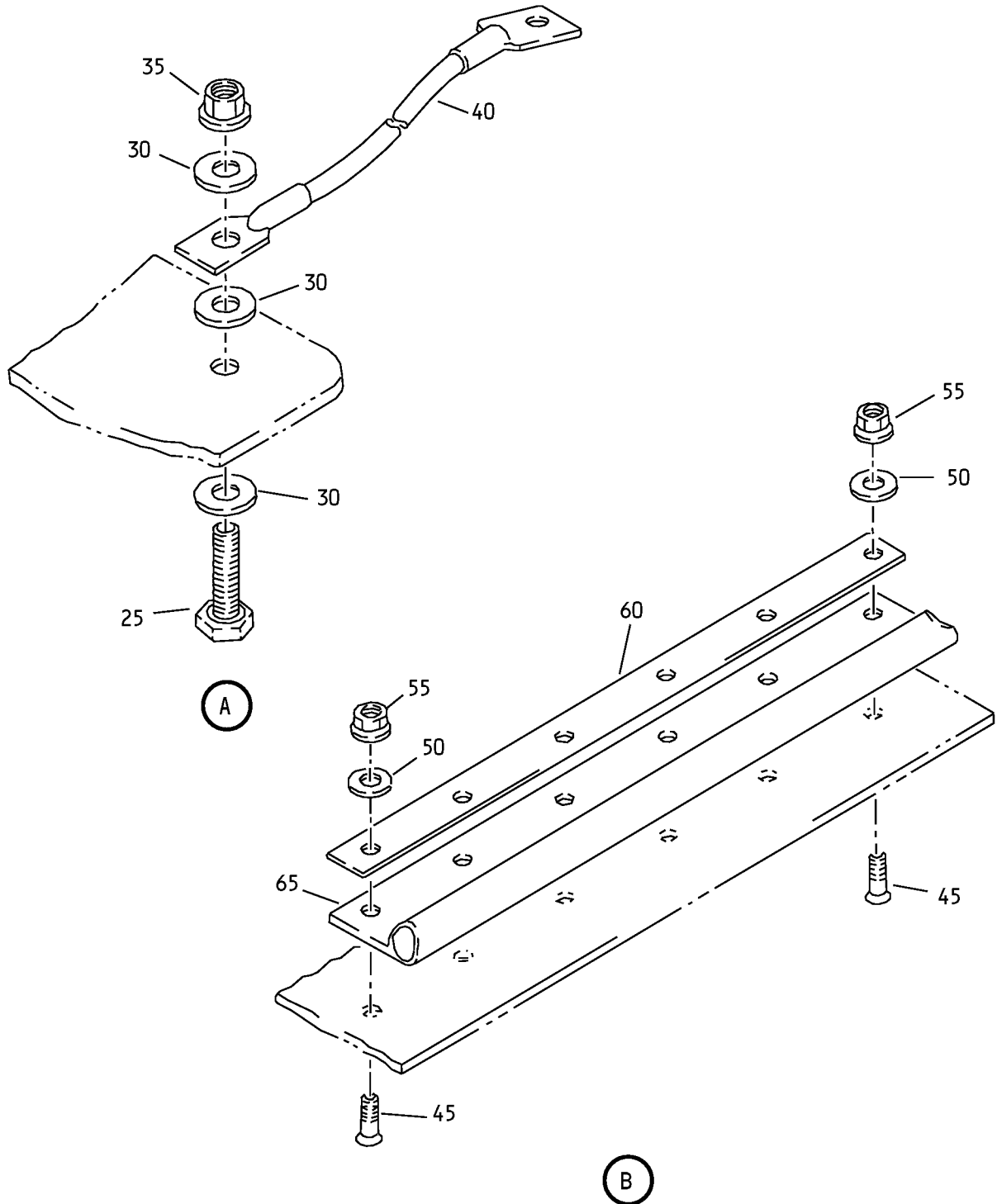
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No. 5 and 8 Flight Spoiler Assembly
IPL Figure 1 (Sheet 2 of 5)

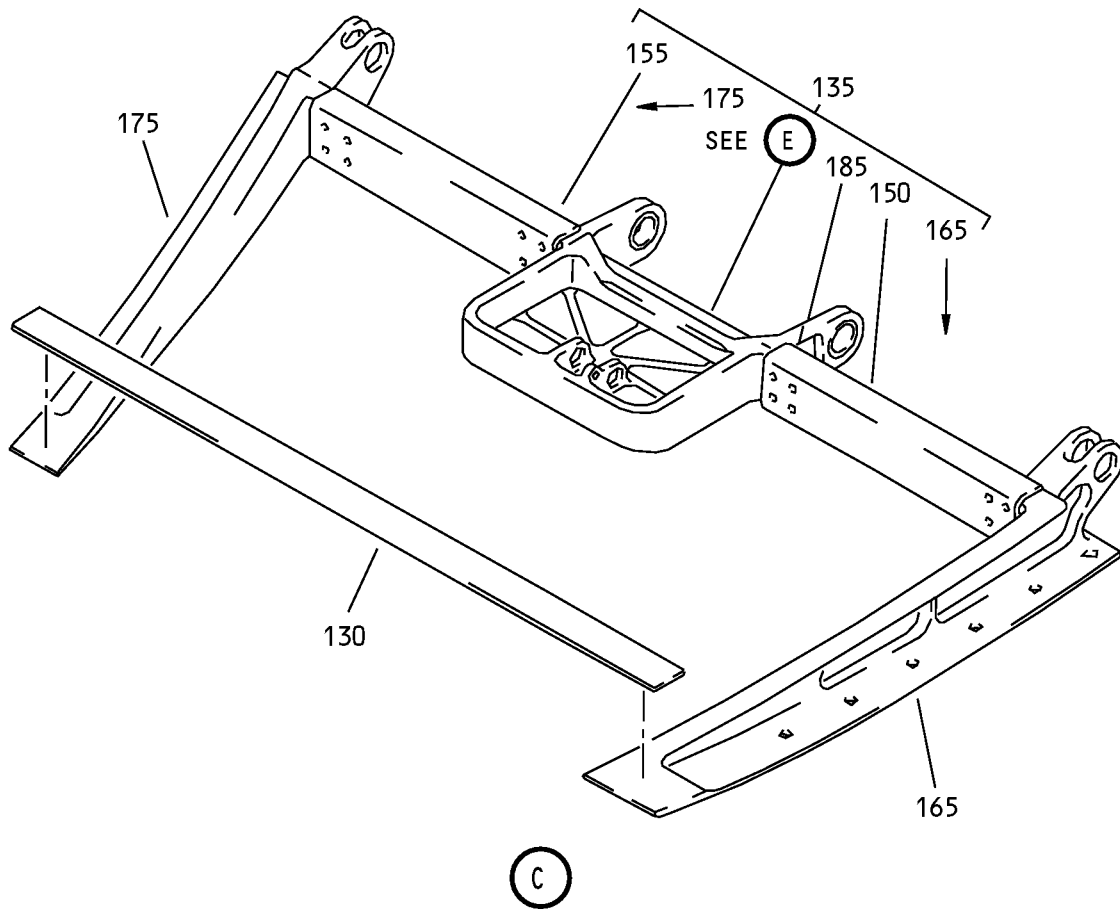
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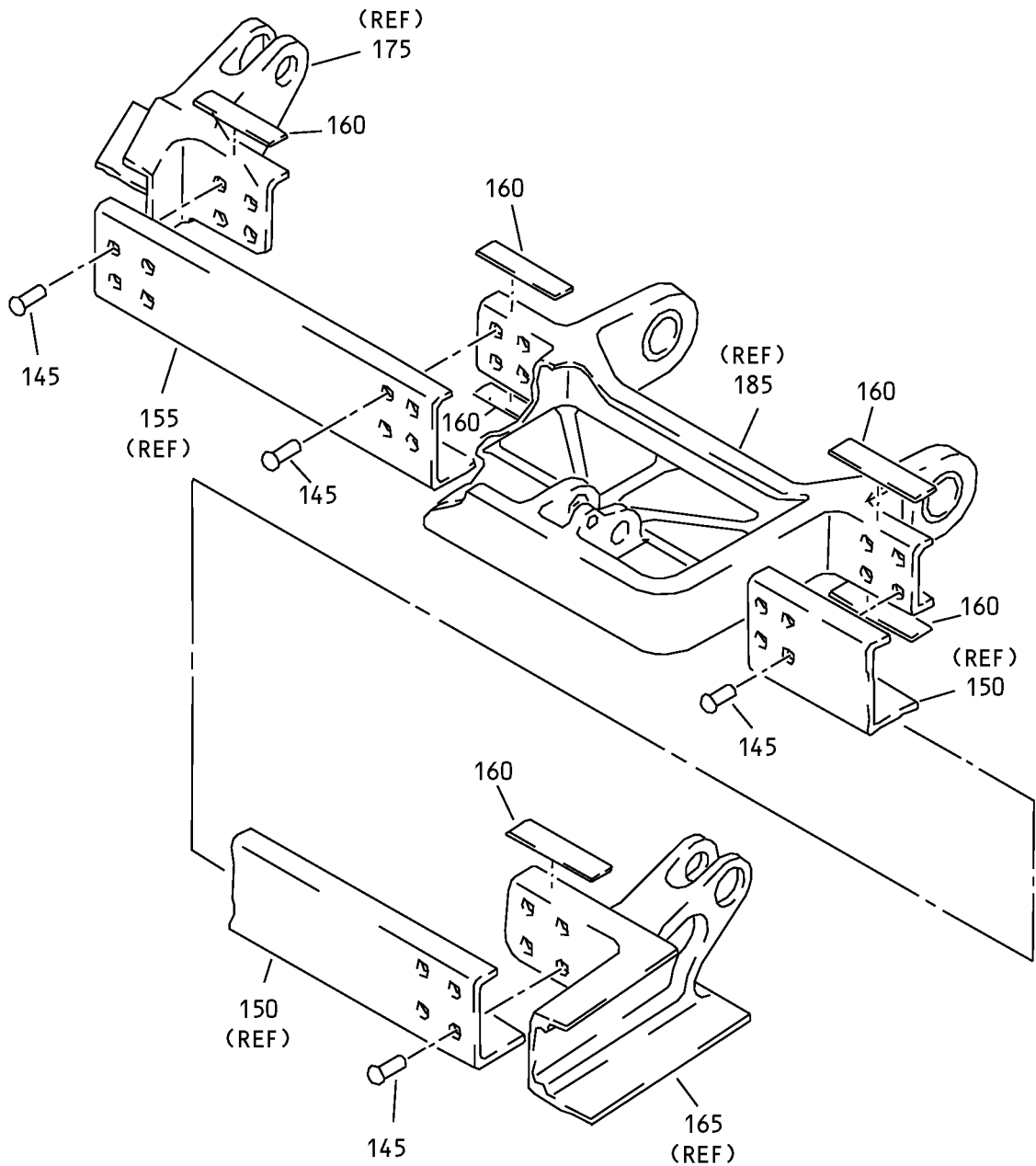
WARNING
HAZARDOUS - AREA

TO PREVENT AUTOMATIC RETRACTION IF HYDRAULIC PRESSURE IS APPLIED OR IF ELECTRICAL POWER IS LOST, DEACTIVATE SPOILER/SPEEDBRAKE CONTROL SYSTEM PER MAINTENANCE MANUAL SECTION 27-61-00 BEFORE ENTERING AREA TO PERFORM ANY MAINTENANCE OPERATION.

D

No. 5 and 8 Flight Spoiler Assembly
IPL Figure 1 (Sheet 3 of 5)

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(E)

No. 5 and 8 Flight Spoiler Assembly
IPL Figure 1 (Sheet 4 of 5)

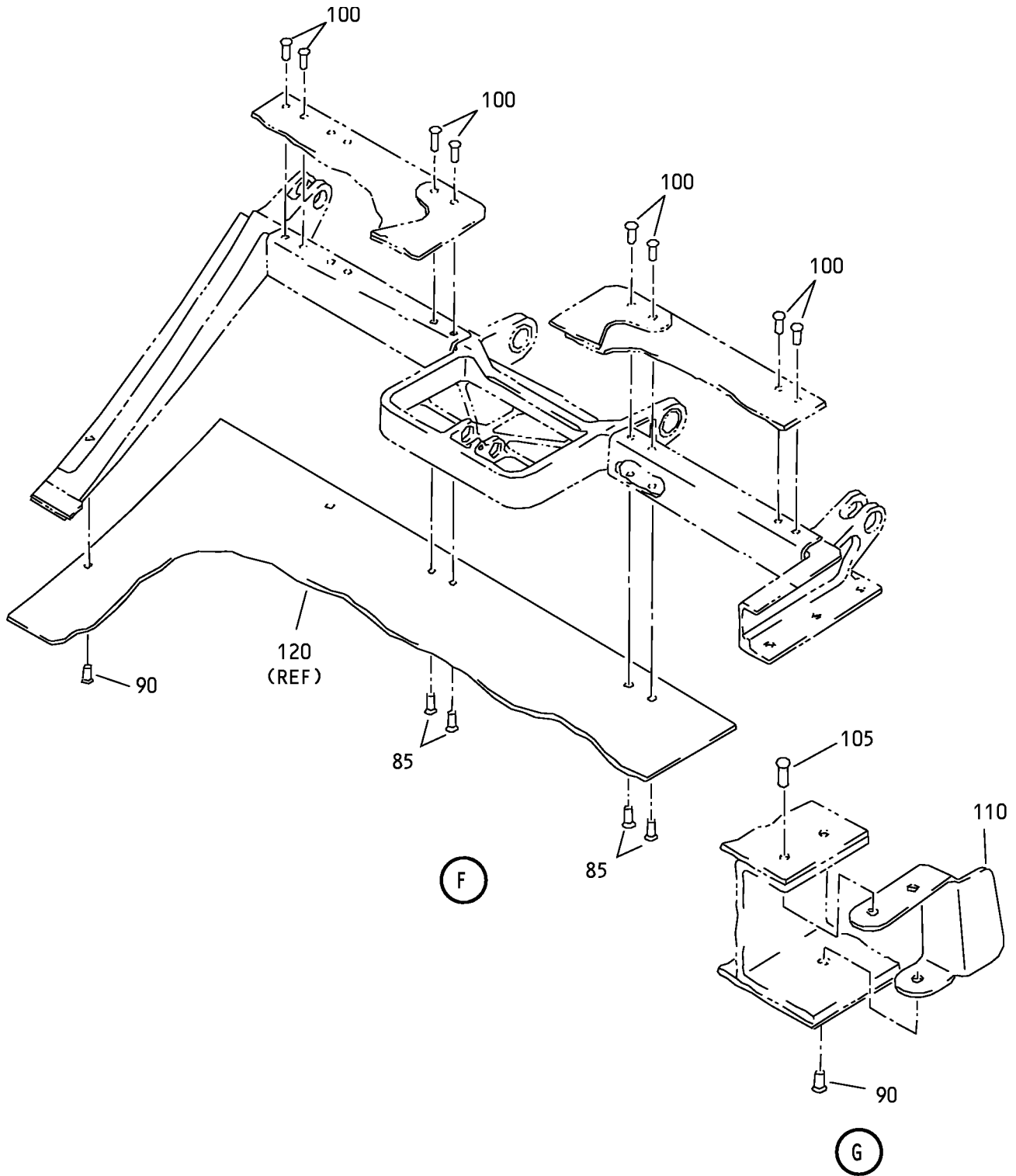
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F79286 S00041007857_V2

No. 5 and 8 Flight Spoiler Assembly
IPL Figure 1 (Sheet 5 of 5)

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-1A	113A4500-1									A	RF
-1B	113A4500-3									C	RF
-5	113A4500-2									B	RF
-5A	113A4500-4									D	RF
10	BACB28AP04P014										2
15	BACB28AT06B014C										2
20	ADW06V301NC										2
25	BACS12GU3K7										2
30	NAS1149D0316J										6
35	H52732-3CD										2
40	940CW20-7										2
45	BACB30VF08K4										6
50	NAS1149DN832J										6
55	BACN10JC08CD										6
60	113A4160-3									A, B	1
-60A	113A4160-11									C, D	1
65	113A4150-11										1
70	BACB28X12M029									A, B	2
-70A	BACB28X12K029									C, D	2

-Item not illustrated

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1- 75	113A4150-5		.	SEAL (MAKE FROM SILICONE RUBBER PER BMS1-57, MAKE FROM 113A4150-7 X 10.0 X 2.0 X 2.0)						A, C	1
-80	113A4150-6		.	SEAL (MAKE FROM SILICONE RUBBER PER BMS1-57, MAKE FROM 113A4150-7 X 10.0 X 2.0 X 2.0)						B, D	1
85	BACR15CE6D		.	RIVET (SIZE DETERMINED ON INST)						A, B	4
-85A	BACR15GF6D7		.	RIVET						C, D	4
90	BACR15CE5D		.	RIVET (SIZE DETERMINED ON INST)						A, B	2
-90A	BACR15GF5D5		.	RIVET						C, D	2
-95	BACR15CE4D			DELETED							
100	BACR15FT6D		.	RIVET (SIZE DETERMINED ON INST)							8
105	BACR15FT5D		.	RIVET (SIZE DETERMINED ON INST)						A, B	2
-105A	BACR15FT5D5C		.	RIVET						C, D	2
110	113A4170-1		.	CLIP						A, C	1
-115	113A4170-2		.	CLIP						B, D	1
120	113A4510-1		.	BOND ASSY-PNL						A, C	1
-125	113A4510-2		.	BOND ASSY-PNL						B, D	1
130	113A4114-5		..	STRIP-RUB							1
135	113A4510-3		..	FRAME ASSY						A, C	1
-140	113A4510-4		..	FRAME ASSY						B, D	1
145	BACR15FT6D		...	RIVET (SIZE DETERMINED ON INST)							16
150	113A4111-7		...	CHANNEL							1
155	113A4111-9		...	CHANNEL							1
160	BACS40R008B018F		...	SHIM							AR
165	113A4112-17		...	RIB-END						A, C	1
-170	113A4112-18		...	RIB-END						B, D	1
175	113A4112-19		...	RIB-END						A, C	1

-Item not Illustrated

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FIG/ ITEM	PART NUMBER	AIRLINE PART NUMBER	NOMENCLATURE							USAGE CODE	UNITS PER ASSY
			1	2	3	4	5	6	7		
1-											
-180	113A4112-20		.	.	.	RIB- END				B, D	1
185	113A4520-1		.	.	.	FITTING- CTR				A, C	1
-190	113A4520-2		.	.	.	FITTING- CTR				B, D	1
195	BACR15BB3D		.			RIVET (SIZE DETERMINED ON INST)				A, B	2
-195A	BACR15BB3D4C		.			RIVET				C, D	2
200	MS27253-1		.			PLATE-IDENT					1
205	BAC27NCT0217		.			MARKER-ALUMINUM FOIL					1

-Item not Illustrated

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