



TO: ALL HOLDERS OF LEADING EDGE AND TRAILING EDGE FLAPS CONTROL ASSEMBLY  
OVERHAUL MANUAL, 27-54-12

REVISION NO. 11, DATED JUL 1/01

HIGHLIGHTS

DESCRIPTION OF CHANGE	TOPICS AFFECTED												
	D & O	D / A s s y	C l e a n i n g	I n s p / C h k	R e p a i r	A s s y	F / C	T e s t	T / S h o o t i n g	S / T o o l s	S t o r a g e	I P L	L / O v e r h a u l
Added SB 27-1241												X	
Added information to clarify the hole on crank assembly (565) for bolt (535) installation						X						X	
Edited without technical change		X											

# LEADING EDGE AND TRAILING EDGE FLAPS CONTROL ASSEMBLY

## 27-54-12

BOEING P/N 65-51602-39, -43, -47, -49, -51, -53

AIRLINE P/N

THE FOLLOWING DIRECTIVES APPLY TO THIS SUBJECT:

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
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OVERHAUL MANUAL

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\*[1] Special instructions not required. Use standard industry practices and information contained in 20-30-01 and 20-30-03.

\*[2] Special instructions not required. Use standard industry practices and information contained in 20-44-02.

LEADING EDGE AND TRAILING EDGE FLAPS CONTROL ASSEMBLY

1. Description and Operation

- A. The LE and TE flaps control assembly consists of a control assembly, a TE flap control valve assembly, and a LE flap control valve assembly.
- B. Inside the control assembly is an input and follow-up linkage for the TE flaps control valve and a control linkage for the LE flaps control valve. A shaft supported by ball bearings in the housing carries five switch cams and two valve cams. The shaft end which protrudes through the housing carries a cable drum. Attached to the housing is a cable guard protecting the drum.
- C. A mechanical input positions the TE flap control valve which directs (available) hydraulic pressure to the TE flap actuators. Motion of the TE flaps, through a follow-up system, limits amount of flap actuation and sequentially positions the LE flap control valve, which directs hydraulic pressure to the LE flaps for coordinated action with the TE flaps. Integral cams, in turn, actuate appropriate switches (not part of assembly) to indicate system direction and to signal improper conditions for flap positioning.
- D. Leading particulars (approximate)
  - Length (overall) -- 16 inches
  - Height (overall) -- 13 inches
  - Width (overall) -- 9.5 inches
  - Weight (approx) -- 20 pounds

2. Disassembly (Fig. 3)

- A. Remove screws (60), cover assembly (65) and pan assembly (80).
- B. Remove bolts (95), washers (100) and drum guard (105).
- C. Remove nut (150), washer (155), eyebolt (160) and spring (115).
- D. Remove nut (120), washer (125), bolt (130). Remove bolts (1), washers (5) and valve assembly (10). Remove parts (15 thru 30) from valve assembly (35) as required.
- E. Remove plugs (165) then remove parts (170 thru 185). Remove bolts (40), washers (45) and valve assembly (50) from housing assembly (690).

NOTE: Refer to 27-53-12 for overhaul procedures for valve assembly (50).

- F. Remove nut (190), washer (195) and bolt (200). Loosen nut (210) and separate link assembly (135) from bearing (215).

NOTE: Do not remove bearing (140) from link (135) unless necessary for repair or replacement.

- G. Remove parts (220 thru 300).

NOTE: Do not remove bushings (305) from lever (310) unless necessary for repair or replacement.

- H. Remove parts (315 thru 335) and follower arm assemblies (365, 410).

- I. Remove nut (350), washer (355), bearing (360), tee (362), bearing (340) and bushing (345) from arm assembly (365).

NOTE: Do not remove bearing (370) from arm (375) unless necessary for repair or replacement.

- J. Remove nut (395), washer (400), bearings (380, 405), spacer (385, 390) from arm assembly (410).

NOTE: Do not remove bearing (415) from arm (420) unless necessary for repair or replacement.

- K. Remove cotter pin (425), nut (430) from shaft (510). Remove parts (435 thru 470) and remove shaft (510) from housing assembly (690). Remove parts (475 thru 490).

- L. Remove drum (495) from shaft (510). Remove pin (500) and plug (505).

- M. Remove screws (515), plate (520) and input crank (565) from housing assembly (690). Remove parts (525 thru 535) and separate link assembly (540) from input crank (565).

NOTE: Do not disassemble link assembly (540) or crank assembly (565) unless necessary for repair or replacement.

- N. Remove parts (590 thru 635, 685) from housing assembly (690).

NOTE: Do not disassemble housing assembly (690) unless necessary for repair or replacement.



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3. Inspection/Check

- A. Check all parts for obvious defects in accordance with standard industry practices.
- B. Penetrant check per 20-20-02 -- guard (105), tee (362), links (145, 270, 550), sleeve (255), lever (310), arms (375, 420), drum (495), shaft (510), crank (585) and housing (715).
- C. Magnetic particle check per 20-20-01 -- cams (475, 485), pin (500).
- D. Check springs (110, 115) per Fig. 1.

Spring Data	Spring (110)	Spring (115)
Free length inside hooks (inches) (Suggested specification)	3.24	3.07
Initial tension (pounds)	3.40 - 4.60	3.50 - 4.50
Load at 4.12 inches (pounds)	9 - 11	
Load at 4.82 inches (pounds)		18 - 22
Load at 5.85 inches (pounds)	20 - 24	
Maximum extended length Without permanent set (inches)		5.25

Spring Check Data  
Figure 1

**OVERHAUL MANUAL**4. Repair

A. Repair minor defects in accordance with standard industry practices.

B. Refinish

NOTE: Refer to 20-30-02 for stripping of protective finish and to 20-41-01 for decoding Boeing finish codes.

- (1) Cover (75), pan (80, 90, 65-51617-2 only), tee (362), spacers (440, 470, 480, 490), plates (520, 615) -- Alodize or chromic acid anodize and apply 1 coat of BMS 10-11, type 1, (F-2.30) all over. Material: Al alloy.
- (2) Guard (105), lever (310), arm (375) -- Chromic or sulfuric acid anodize (F-17.05) and apply 1 coat of primer, BMS 10-11, type 1 (F-20.02) all over. Material: Al alloy.
- (3) Springs (110, 115, 290) -- Cadmium plate and apply 1 coat of primer, BMS 10-11, type 1 (SRF-1.92) all over. Material: Steel wire per QQ-W-470.
- (4) Link (145) -- Passivate (F-17.09) all over. Material: 17-4PH CRES, 150-170 ksi.
- (5) Sleeve (255) -- Passivate (F-8.07) all over. Material: 303 CRES.
- (6) Link (270) -- Chromic or sulfuric acid anodize (F-17.05) and apply 1 coat of primer, BMS 10-11, type 1 (F-20.02) except omit primer on threads. Material: Al alloy.
- (7) Cams (445 thru 465, 65-51618-1, -4, -5, -7, -9, -10) -- Hard anodize (F-2.204) on peripheral surface only. Flash hard coat 0.0002-0.0004 in. thick and seal with sodium dichromate per MIL-A-8625 all other surfaces. Material: Al alloy.
- (8) Cam (467, 65-51618-12) -- Hard anodize (F-17.06) on peripheral surface only. Flash hard coat 0.0002-0.0004 in. thick and seal with sodium dichromate per MIL-A-8625 all other surfaces. Material: Al alloy.
- (9) Cams (445 thru 467, 65-51618-13 thru -19) -- Cadmium plate 0.0002 to 0.0004 inch thick (F-15.02) all over. Material: CRES.
- (10) Cam (475) -- Passivate (F-8.07) all over and cadmium plate (F-15.02) on splines and both sides within 0.75 in. radius only. Material: 15-5 CRES, 180-200 ksi.



- (11) Cam (485) -- Passivate (F-8.07) all over and cadmium plate (F-1.1926) on splines and both sides within 0.75 in. radius only. Material: 17-7PH or 17-4PH CRES, 180-200 ksi.
- (12) Drum (495) -- Chromic or sulfuric acid anodize (F-17.05) and apply 1 coat of primer, BMS 10-11, type 1 (F-20.02) all over except omit primer on splines. Material: Al alloy.
- (13) Pin (500) -- Cadmium plate (F-1.1926) all over. Material: 4340 steel, 125-145 ksi.
- (14) Plug (505) -- Chromic acid anodize (F-2.26) all over and apply 1 coat of primer, BMS 10-11, type 1 (SRF-12.205) on all external surfaces. Material: Al alloy.
- (15) Shaft (510) -- Chromic acid anodize (F-2.26) all over and apply 1 coat of primer, BMS 10-11, type 1 (SRF-12.205) except no primer on splines and threads. Material: Al alloy.
- (16) Link (550) -- Chromic acid anodize and apply 1 coat of primer, BMS 10-11, type 1 (F-18.13) except omit primer in bores for bearings. Material: Al alloy.
- (17) Crank (585) -- Chromic or sulfuric acid anodize (F-17.05) and apply 1 coat of primer, BMS 10-11, type 1 (F-20.02) except no primer on splines, bearing seats or 0.2495-0.2505 and 0.257 inch diameter holes. Material: Al alloy.
- (18) Housing (715) -- Chromic or sulfuric acid anodize (F-17.05) and apply 1 coat of primer, BMS 10-11, type 1 (F-20.02) except omit primer in holes. Material: Al alloy.
- (19) Arm (420) -- Anodize (F-17.05) and apply primer (F-20.02) except omit primer on bearing seat.
- (20) Cover Pan (80, 65-51617-4 and 65C32708-1) -- Chemical treat and apply one coat of primer, BMS 10-11, type 1 (F-18.06) all over. Material: Al alloy.

C. Replacement

- (1) Bearing (140) - Install replacement bearing with BMS 5-95 sealant and point stake per 20-50-03.
- (2) Bearings (370, 415, 545) -- Install replacement bearings with BMS 5-95 sealant and roller swage per 20-50-03.
- (3) Bearing (575) -- Install replacement bearing with BMS 5-95 sealant and swage ring (570) per 20-50-03.

- (4) Nameplates (655 thru 682) -- Install replacement nameplates and bond per 20-50-12, type 70. Fay surface coverage of 100 percent and edge squeeze-out is required. If required, steel stamps assembly number and serial number on nameplate (682) as given in SOPM 20-50-10.
- (5) Bushings (305) -- Install replacement bushing as given in SOPM 20-50-03 and machine bushing I.D. thru to 0.4370-0.4380 inch diameter.

5. Assembly (Fig. 3)

- A. Install retainer assembly (635) in housing assembly (690) and secure with bolts (625) and washers (630) using BMS 5-95 sealant.
- B. Install bearings (620) in housing assembly (690) with sealant. Install retainer (615) and secure with screws (605) and washers (610). Install screws (605) and washers (610) with sealant.
- C. Install bolt (590), washer (595) and spacer (600) in housing assembly (690).
- D. Assemble the link assembly (540) on the crank assembly (565). Crank assembly (565) has two holes. The 0.2495-0.2505 inch diameter hole is used for bolt (535), the 0.323-0.327 inch diameter hole is for a rig pin that is used in rigging the valve assemblies (10) and (50). Make sure that bolt (535) is installed into the correct hole. Install bolt (535), washer (530) and nut (525) with BMS 5-95 sealant.

**CAUTION: DO NOT TIGHTEN BOLT (555) OR DAMAGE TO INPUT CRANK ASSEMBLY (565) WILL RESULT.**

- E. Install bolt (555), washer (560) on input crank assembly (565). Do not tighten bolt (555). Assemble input crank assembly (565) in housing assembly (690) and secure with plate (520) and screws (515).
- F. Assemble shaft (510) and cams (475, 485).
  - (1) Install plug (505) and join (500) on shaft (510) with sealant.
  - (2) Coat splines of drum (495) with grease BMS 3-24 and install drum on shaft (510).
  - (3) Apply a coat of grease MIL-G-23827 to cam surface and grease BMS 3-24 to splines of cams (475, 485). Position spacers (472, 480, 490) and cams (475, 485) in housing assembly (690) and install shaft (510) with attached drum (495).
  - (4) Apply a coat of grease MIL-G-23827 to cam surface and grease BMS 3-24 to splines of cams (445 thru 467). Install spacer (470) and cams (445 thru 467) on shaft (510). Install cams (445 thru 467) with identification numeral facing outward.
  - (5) Install spacer (440), washers (435) and nut (430) on shaft (510). Install nut (430) with grease BMS 3-24 and tighten to 250-400 lb-ins. Install cotter pin (425).

**NOTE: Use washers AN960D1616L as required with washers (435) to facilitate installation of cotter pin.**

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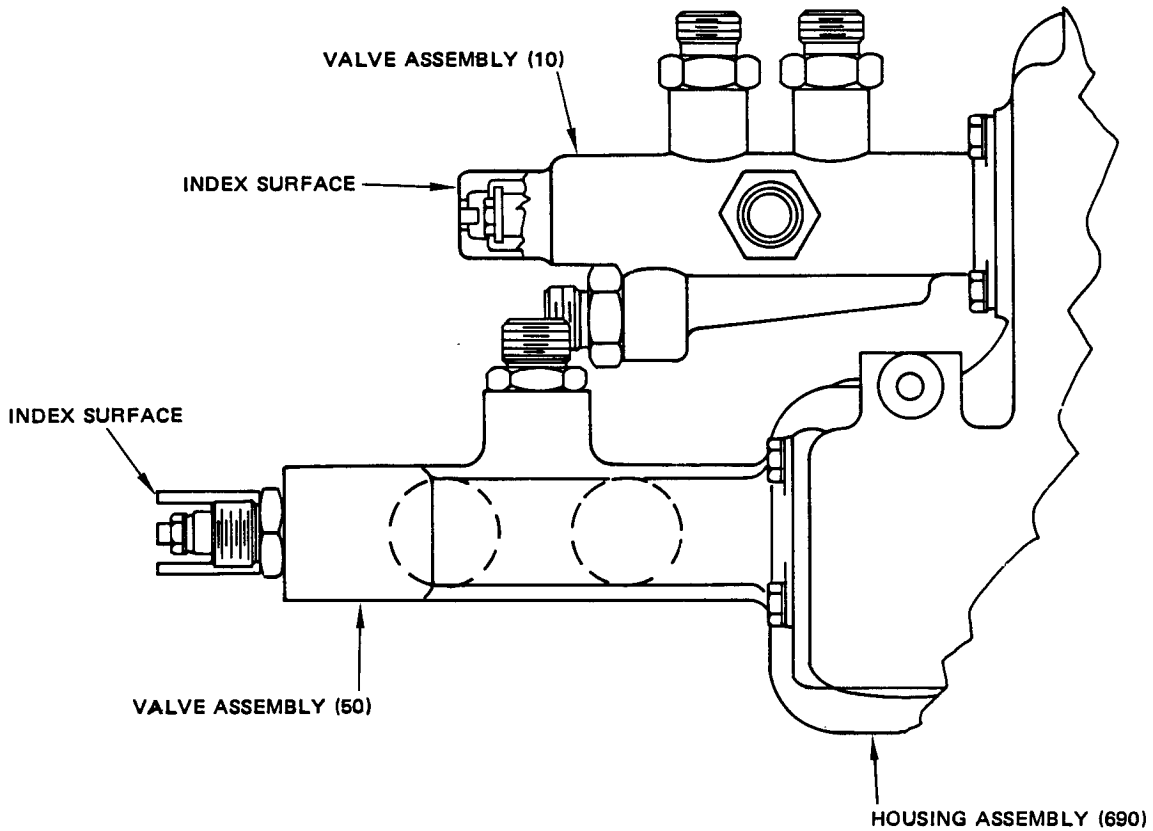
- G. Lightly lubricate bearing (405) with grease, MIL-G-23827 and install on follower arm assembly (410). Secure bearing with washer (400) and nut (395). Install bearings (380) and spacers (385, 390).
  - H. Lightly lubricate bearing (360) with grease, MIL-G-23827 and install on follower arm assembly (365). Position tee (362) on follower arm assembly (365) and install washer (355) and nut (350) to secure bearing (360). Install bearing (340) and spacer (345).
  - I. Position follower arm assemblies (365, 410) in housing assembly (690) and secure with bolt (325), washers (320), bushings (330, 335) and nut (315).
  - J. Install bushings (295), leaf spring (290), eyebolt (285), washers (280) and nut (275) on summing lever assembly (310). Install bushings (295) with grease, MIL-G-23827.
  - K. Assemble nut (260), sleeve (255), washer (245), nut (250) and bearing (240) on link (270).
- NOTE: Do not tighten nuts (260, 250).
- L. Position link (270) on follower arm assembly (410) and secure with bushing (265), bolt (235), washer (225) and nut (220).
  - M. Position summing lever assembly (300) on link assembly (540) and secure with parts (220, 225, 230).
  - N. Lightly coat packings (30) with hydraulic fluid, BMS 3-11 and install on reducers (20, 25) and union or reducer (15). Install parts (15, 20, 25) on valve assembly (35).
  - O. Install valve assemblies (10, 50) on housing assembly (690) and secure with bolts (1, 40) and washers (5, 45). Install bolts (1, 40) and washers (5, 45) with sealant.
  - P. Connect valve assembly (50) to summing lever assembly (300) with bolt (185), washer (180), nut (175), and cotter pin (170).
  - Q. Assemble bearing (215), washer (205), nut (210) on link assembly (135).
  - R. Install link assembly (135) on valve assembly (10) and secure with bolt (130), washer (125) and nut (120) using BMS 5-95 sealant.
  - S. Rigging valve assemblies (10, 50).
    - (1) Install 0.309-0.311 in. diameter rigging pin thru hole in housing assembly (690) to secure input crank assembly (565).
    - (2) Install a rigging pin (MS20392 or equivalent) thru cam (485) and follower arm assembly (410).

- (3) Rigging procedures for valve assembly (50).
    - (a) Turn bearing (240) in 180 degree increments. Install bolt (235) and check that slide of valve assembly (50) aligns with index surfaces (Fig. 2).
    - (b) When approximate position of slide is reached, install parts (220, 225, 230) thru summing lever assembly (300) and bearing (240).
    - (c) Rotate sleeve (255) to obtain exact position of slide then tighten nuts (250, 260).
  - (4) Rigging procedures for valve assembly (10)
    - (a) Rotate bearing (215) in 180 degrees increment. Install bolt (200) and check that slide aligns with index surface shown in Fig. 2.
    - (b) When proper slide position is obtained, attach bearing (215) to follower arm assembly (365) with bolt (200), washer (195) and nut (190). Use BMS 5-95 sealant with nut (190).
  - (5) Remove rigging pins.
- T. Install eyebolt (160) in housing assembly (690) and secure with washer (155) and nut (150). Install springs (110, 115).
- U. Install guard (105) and secure with bolts (95) and washers (100).
- V. Check that unit operates smoothly when cable drum (495) is rotated in both directions.
- W. Lockwire the following parts with MS20995NC32 using double twist method per 20-50-02.
- (1) Nut (260) to link (270).
  - (2) Nut (250) to lockwasher (245).
  - (3) Nut (210) to lockwasher (205).
- X. When using the optional cover assembly (65, 69-38188-1), or cover pan assembly (80, 65-51617-1), coat surface of gaskets (70, 85) with grease, MIL-G-23827. Install cover (65) and cover pan (80) on housing assembly (690) and secure with screws (60). Install screws (60) with sealant.
- Y. Install plugs (165, 685) on housing assembly (690) and fillet seal seam around plugs with sealant.

Z. Materials.

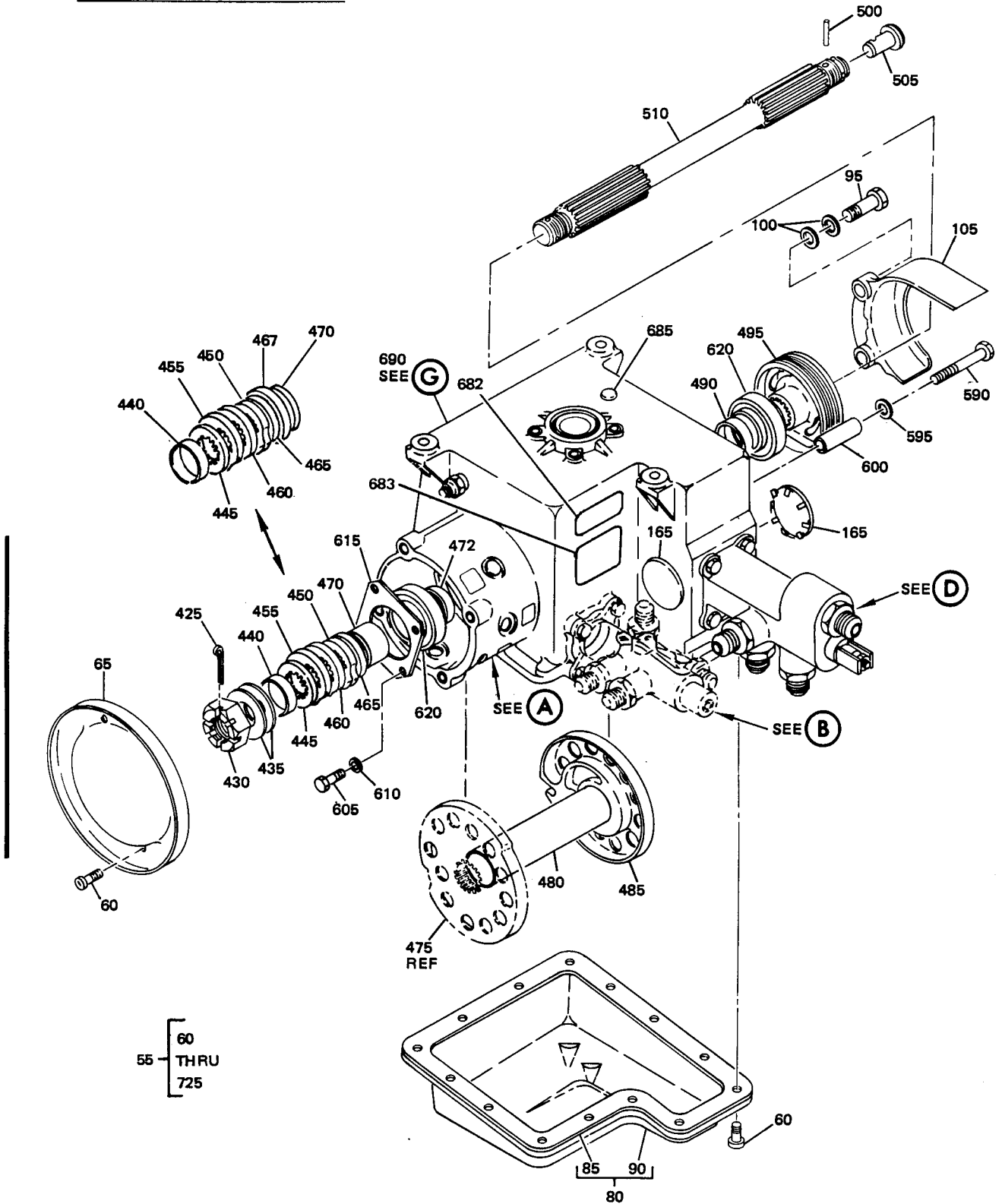
NOTE: Equivalent substitutes may be used.

- (1) Grease -- MIL-G-23827 (Ref 20-60-03).
- (2) Grease -- BMS 3-24 (Ref 20-60-03).
- (3) Hydraulic fluid -- BMS 3-11 (Optional: Skydrol assembly lube MCS 352) (Ref 20-60-03).
- (4) Sealant -- BMS 5-95 (Ref 20-60-04).
- (5) Lockwire -- MS20995NC32.



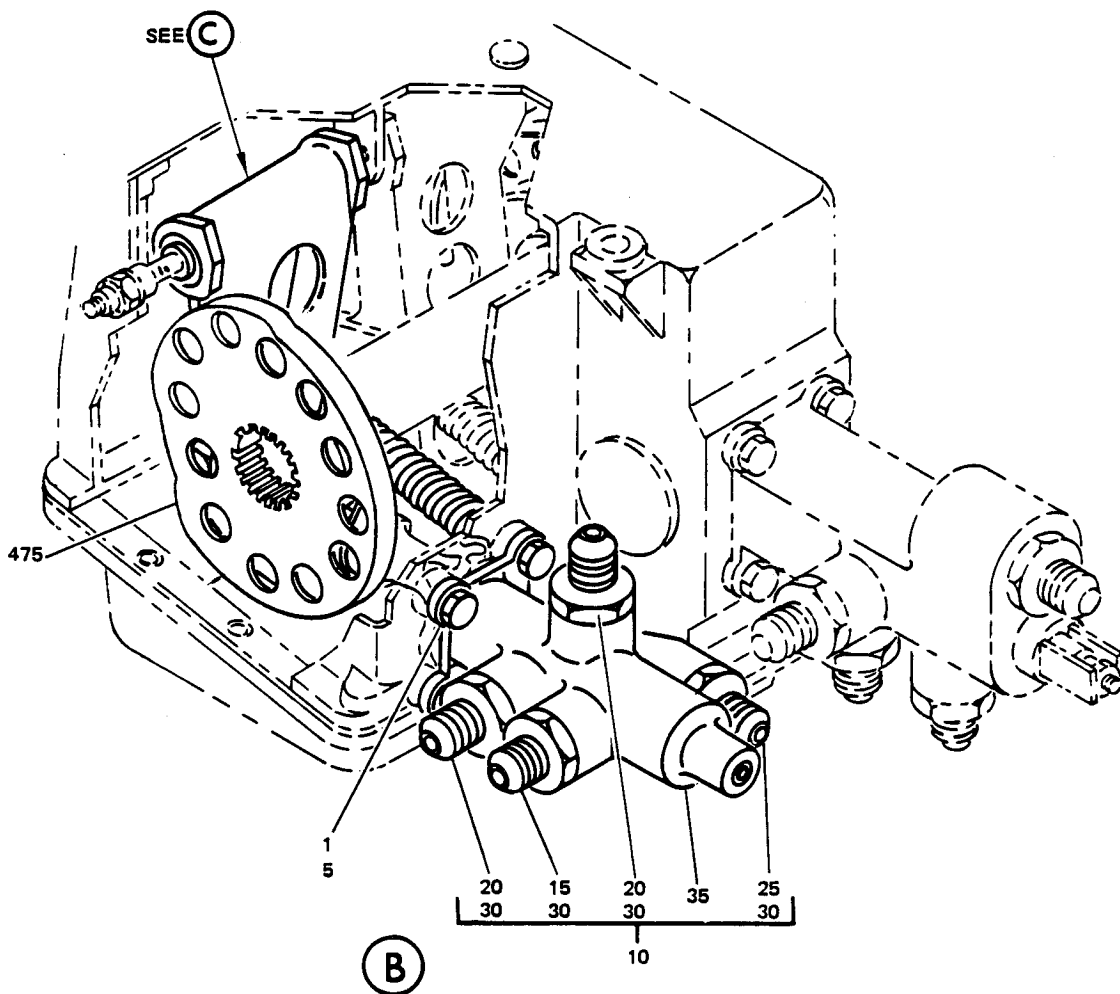
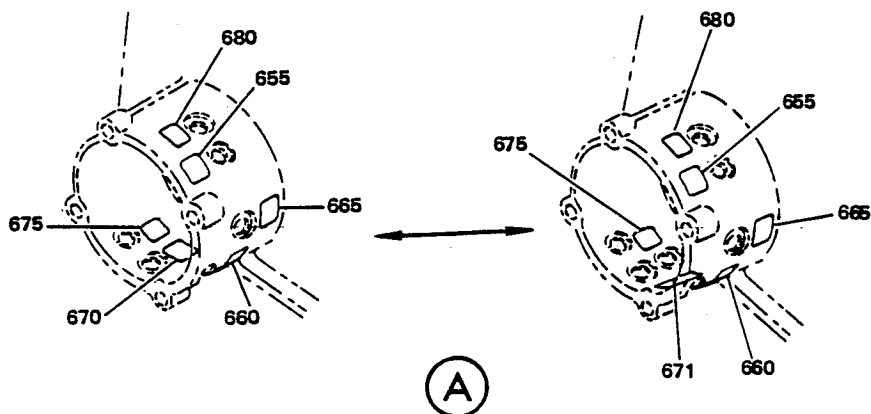
Rigging Diagram  
Figure 2

6. ILLUSTRATED PARTS LIST

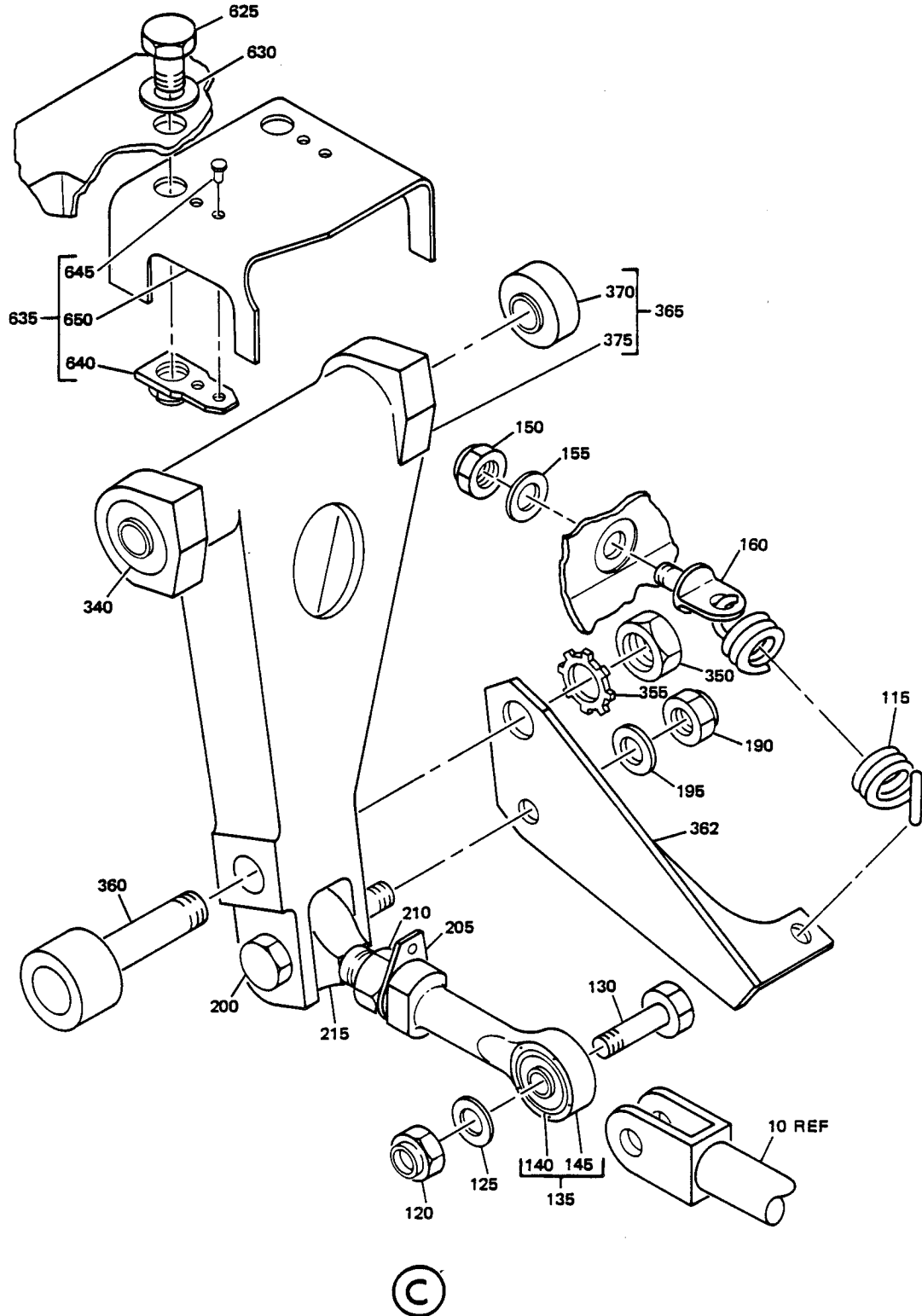


Leading Edge and Trailing Edge Flaps Control Assembly  
 Figure 3 (Sheet 1)

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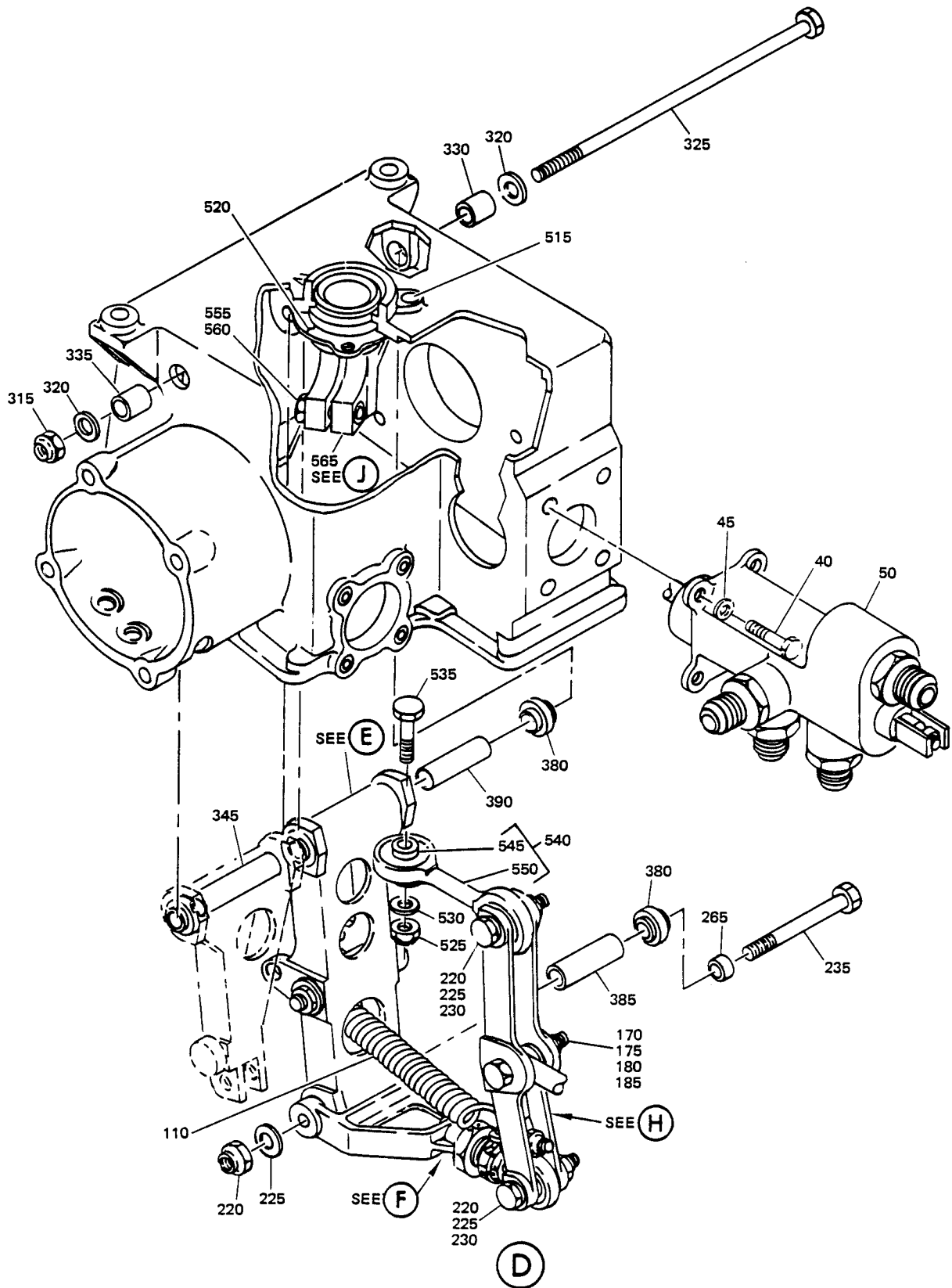
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Leading Edge and Trailing Edge Flaps Control Assembly  
Figure 3 (Sheet 3)

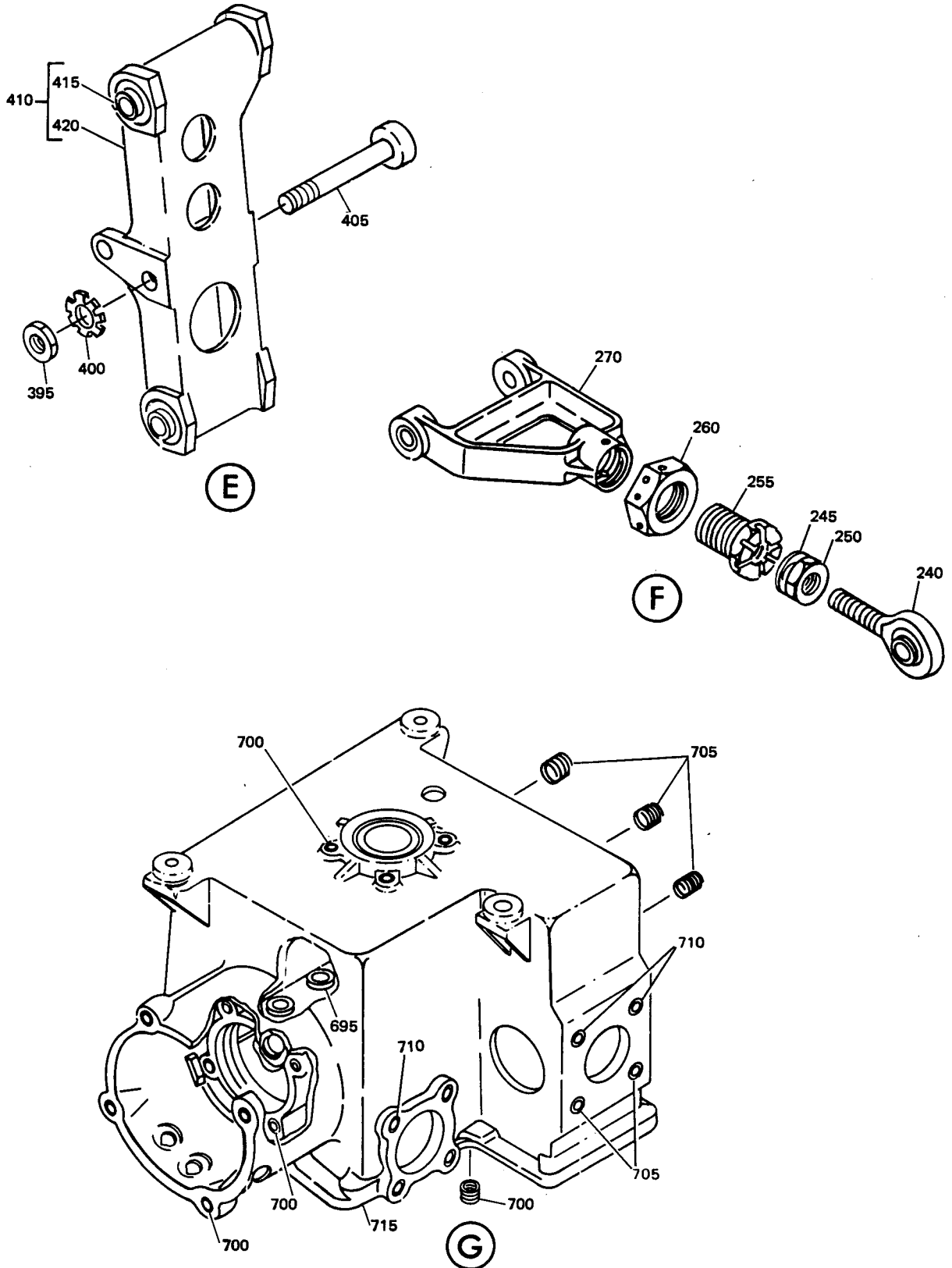


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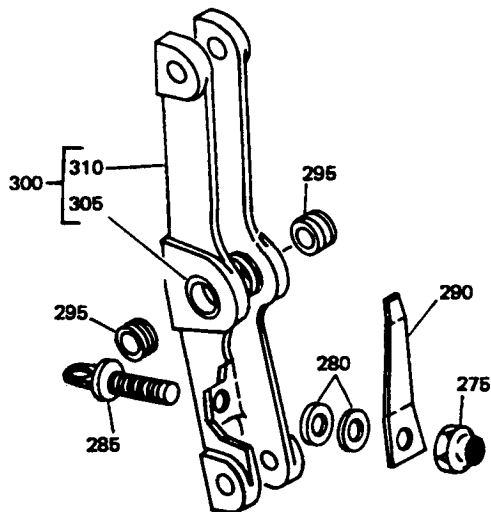


Leading Edge and Trailing Edge Flaps Control Assembly  
Figure 3 (Sheet 4)

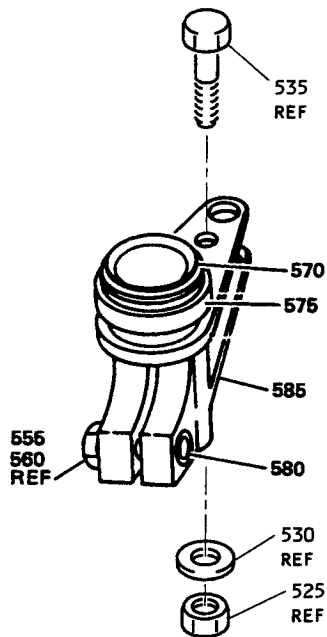
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Leading Edge and Trailing Edge Flaps Control Assembly  
Figure 3 (Sheet 5)



H



J

Leading Edge and Trailing Edge Flaps Control Assembly  
Figure 3 (Sheet 6)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
3-	65-51602-37		DELETED								
	65-51602-39		CONTROL ASSY, LE AND TE FLAP							A	RF
	65-51602-41		DELETED								
	65-51602-43		CONTROL ASSY, LE AND TE FLAPS							B	RF
	65-51602-49		CONTROL ASSY, LE AND TE FLAPS (POST SB 27-1180)(POST SB 27-1241)							C	RF
	65-51602-47		CONTROL ASSY, LE AND TE FLAPS							D	RF
	65-51602-51		CONTROL ASSY, LE AND TE FLAPS (PRE SB 27-1180)(PRE SB 27-1241)							E	RF
	65-51602-53		CONTROL ASSY, LE AND TE FLAPS							F	RF
1	NAS6604-4		. BOLT								4
1	NAS1304-4H		DELETED								
5	AN960D416		. WASHER								4
10	65-51602-38		. VALVE ASSY, LE FLAP CONTROL							B-EF	1
10	65-51602-3		. VALVE ASSY, LE FLAP CONTROL							A	1
15	MS21916-8-6		. . REDUCER							B-EF	1
15	MS21902-6		. . UNION							A	1
20	MS21916-8-6		. . REDUCER								2
25	MS21916D8-6		. . REDUCER								1
30	NAS1612-6		. . PACKING								4
35	4163DA		. . VALVE ASSY, V78062 (BOEING 10-60598-1)								1
40	NAS6604-3		. BOLT								4
40	NAS1304-3H		DELETED								
45	AN960D416		. WASHER								4
50	65-51602-11		. VALVE ASSY (REF 27-53-12)								1
55	65-51602-40		. CONTROL ASSY							A	1
55	65-51602-42		DELETED								
55	65-51602-44		. CONTROL ASSY							B	1
55	65-51602-46		. CONTROL ASSY							D	1
55	65-51602-50		. CONTROL ASSY							C	1
55	65-51602-52		. CONTROL ASSY							E	1
55	65-51602-54		. CONTROL ASSY							F	1
60	NAS1801-3-7		. . SCREW *[1]								AR
65	69-38188-2		. . COVER								1
65	69-38188-1		DELETED								
70	69-38188-3		DELETED								
75	69-38188-2		DELETED								
80	65C32708-1		. . PAN, COVER								1
80	65-51617-2		. . PAN, COVER (OPT)							A	1
80	65-51617-4		. . PAN, COVER (OPT)								1
80	65-51617-1		. . PAN ASSY, COVER (OPT)							A	1
85	65-51617-3		. . . GASKET (USED ON 65-51617-1)								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
3-90	65-51617-2		.	.	.	PAN, COVER (USED ON 65-51617-1)				1		
95	NAS1304-7		.	.	DELETED							
95	NAS1304-5		.	.	BOLT							2
100	AN960D416		.	.	DELETED							
100	AN960D416L		.	.	WASHER							2
105	65-51615-1		.	.	GUARD, DRUM							1
110	69-38184-1		.	.	SPRING							1
115	69-57953-1		.	.	SPRING							1
120	BACN10JC4		.	.	NUT							1
125	AN960D416		.	.	WASHER							1
130	NAS1104-8		.	.	BOLT							1
135	69-38185-1		.	.	LINK ASSY							1
140	BACB10AC4A		.	.	.	BEARING					1	
145	69-38185-2		.	.	.	LINK					1	
150	BACN10JC4		.	.	NUT							1
155	AN960PD416		.	.	WASHER							1
160	AN43B4		.	.	EYEBOLT							1
165	BACP20B33		.	.	PLUG							2
170	MS24665-134		.	.	PIN, COTTER							1
175	AN320-4		.	.	NUT							1
180	AN960-416L		.	.	WASHER							2
185	NAS1104-19D		.	.	BOLT							1
190	NAS679A4W		.	.	NUT							1
195	AN960D416L		.	.	WASHER							1
200	NAS1104-15		.	.	BOLT							1
205	NAS513-6		.	.	LOCKWASHER							1
210	NAS509-6		.	.	NUT							1
215	BACB10A446H		.	.	BEARING							1
220	NAS679A4W		.	.	NUT							3
225	AN960D416		.	.	WASHER							3
230	NAS1104-18		.	.	BOLT							2
235	NAS1104-44		.	.	BOLT							1
240	BACB10A446H		.	.	BEARING							1
245	NAS513-6		.	.	LOCKWASHER							1
250	NAS509-6		.	.	NUT							1
255	69-38187-2		.	.	SLEEVE							1
255	69-38187-1		.	.	SLEEVE (OPT)						A-D	1
260	NAS509-10		.	.	NUT							1
265	NAS75-4-008		.	.	BUSHING							1
270	65-51611-4		.	.	LINK, TE CAM							1
275	NAS679A3W		.	.	NUT							1
280	AN960D10		.	.	WASHER							2
285	AN42B5A		.	.	EYEBOLT							1
290	69-38193-1		.	.	SPRING, LEAF							1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
3-295	NAS73-4-006		.	.								2
300	65-51616-6		.	.								1
305	NAS538B7P032		.	.	.							2
310	65-51616-7		.	.	.							1
315	NAS679A4W		.	.								1
320	AN960-416L		.	.								2
325	NAS6604-118		.	.								1
330	NAS73-4E007		.	.								1
335	NAS73-4-005		.	.								1
340	BACB10A661		.	.								1
345	NAS43DD4-174		.	.								1
350	AN316-6R		.	.								1
355	MS27111-3		.	.								1
360	BACB10AF6K16 HS		.	.						ABC		1
360	BACB10AF6K16H		.	.						ABC		1
360	BACB10B60TK11		.	.						ABC		1
360	BACB10FK6K16 HS		.	.						DEF		1
362	69-57951-1		.	.								1
365	65-51607-6		.	.								1
370	BACB10A661		.	.	.							1
375	65-51607-7		.	.	.							1
380	BACB10A661		.	.								2
385	NAS43DD4-109		.	.								1
390	NAS43DD4-162		.	.								1
395	AN316-6R		.	.								1
400	MS27111-3		.	.								1
405	BACB10AF6K34H		.	.						ABC		1
405	BACB10B60TK20		.	.						ABC		1
405	BACB10FK6K34 HS		.	.						DEF		1
410	65-51606-4		.	.								1
415	BACB10A661		.	.	.							1
420	65-51606-5		.	.	.							1
425	MS24665-360		.	.								1
430	AN320-16		.	.								1
435	AN960D1616		.	.								2
440	69-38191-4		.	.								1
445	65-51618-13		.	.								1
445	65-51618-1		.	.								1
450	65-51618-14		.	.								1
450	65-51618-4		.	.								1
455	65-51618-16		.	.						A		1
455	65-51618-7		.	.						A		1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
3-											
455	65-51618-15		.	.	CAM, SWITCH					BCF	1
455	65-51618-5		.	.	CAM, SWITCH (OPT)					BC	1
455	65-51618-20		.	.	CAM, SWITCH					DE	1
460	65-51618-17		.	.	CAM, SWITCH						1
460	65-51618-9		.	.	CAM, SWITCH (OPT)					ABC	1
465	65-51618-18		.	.	CAM, SWITCH						1
465	65-51618-10		.	.	CAM, SWITCH (OPT)					ABC	1
467	65-51618-19		.	.	CAM, SWITCH					B-EF	1
467	65-51618-12		.	.	CAM, SWITCH (OPT)					BC	1
470	69-38191-6		.	.	SPACER					A	1
470	69-38191-7		.	.	SPACER					B-F	1
472	69-38191-4		.	.	SPACER						1
475	65-51608-8		.	.	CAM, LE VALVE						1
475	65-51608-10		.	.	CAM, LE VALVE (OPT)						1
475	65-51608-12		.	.	CAM, LE VALVE (OPT)						1
480	69-38191-1		.	.	SPACER						1
485	65-51609-4		.	.	CAM, TE VALVE					ABD	1
485	65-51609-5		.	.	CAM, TE VALVE					CEF	1
490	69-38191-3		.	.	SPACER						1
495	65-51613-3		.	.	DRUM						1
500	69-52260-1		.	.	PIN						1
505	69-52259-1		.	.	PLUG						1
510	65-51610-4		.	.	SHAFT						1
510	65-51610-3		.	.	SHAFT (OPT)						1
515	NAS514P1032-7		.	.	SCREW						3
520	69-38190-1		.	.	PLATE						1
525	BACN10JC4		.	.	NUT						1
530	AN960D416		.	.	WASHER						1
535	NAS1104-16		.	.	BOLT						1
540	69-38186-1		.	.	LINK ASSY						1
545	BACB10A561		.	.	BEARING						2
550	69-38186-2		.	.	LINK						1
555	BACB10NE-10				DELETED						
555	BACB30NE4-10		.	.	BOLT						1
560	AN960D416		.	.	WASHER						1
565	65-51612-1		.	.	CRANK ASSY						1
570	BACR12Y84		.	.	RING						1
575	BACB10A30DDH		.	.	BEARING						1
580	MS21209F4-20		.	.	INSERT						1
585	65-51612-2		.	.	CRANK						1
590	NAS1304-21		.	.	BOLT						1
595	AN960D416		.	.	WASHER						1
600	NAS43DD4-84		.	.	SPACER						1
605	NAS1351-3-10		.	.	SCREW						4
605	MS24678-11				DELETED						

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
3-610	AN960D10		.	.	W	A	S	H			4
615	69-38189-1		.	.	P	L	A	T			1
620	BACB10A687		.	.	B	E	A	R	I		2
625	NAS1104-3		.	.	B	O	L	T			2
630	AN960D416		.	.	W	A	S	H			2
635	69-57952-1		.	.	R	E	T	A	I	N	1
640	BACN10LGP45		DELETED								
640	BACN10KB4		.	.	.	N	U	T	P	L	2
645	MS20426D		.	.	.	R	I	V	E	T	4
650	69-57952-2		.	.	.	R	E	T	A	I	1
655	BAC27DCT0430		.	.	N	A	M	E	P	L	1
655	BAC27DCT125		DELETED								
660	BAC27DCT0426		.	.	N	A	M	E	P	L	1
660	BACM10S3ABK		DELETED								
665	BAC27DCT0427		.	.	N	A	M	E	P	L	1
665	BACM10S3ABL		DELETED								
670	BAC27DCT0428		.	.	N	A	M	E	P	L	1
670	BACM10S3ACT		DELETED							A	
671	BAC27DCT0432		.	.	N	A	M	E	P	L	1
671	BAC27DCT0367		DELETED							B-F	
675	BAC27DCT0429		.	.	N	A	M	E	P	L	1
675	BACM10S3ACU		.	.	M	A	R	K	E	R	1
680	BAC27DCT0431		.	.	N	A	M	E	P	L	1
680	BAC27DCT183		.	.	M	A	R	K	E	R	1
682	BAC27DCT0433		.	.	N	A	M	E	P	L	1
683	BAC27DCT0441		.	.	D	E	C	A	L	-	1
685	BACP20B3		.	.	P	L	U	G			1
690	65-51603-7		.	.	H	O	U	S	I	N	1
690	65-51603-10		.	.	H	O	U	S	I	N	1
690	65-51603-12		.	.	H	O	U	S	I	N	1
690	65-51603-16		.	.	H	O	U	S	I	N	1
695	NAS77A3-009P		.	.	.	B	U	S	H	I	1
700	MS21209F1-15		.	.	.	I	N	S	E	R	25
705	MS21209F4-15		.	.	.	I	N	S	E	R	5
710	MS21209F4-20		.	.	.	I	N	S	E	R	6
715	65-51603-8		.	.	.	H	O	U	S	I	1
715	65-51603-9		.	.	.	H	O	U	S	I	1
715	65-51603-11		.	.	.	H	O	U	S	I	1
715	65-51603-14		.	.	.	H	O	U	S	I	1
715	65-51603-13		.	.	.	H	O	U	S	I	1
715	65-51603-15		.	.	.	H	O	U	S	I	1

\*[1] QUANTITY OF 18 NAS1801-3-7 SCREWS OPTIONAL TO QUANTITY OF 9 NAS1801-3-7 SCREWS (ONLY 5 SCREWS REQUIRED ON PREFERRED PART 65C32708 COVER PAN VERSUS 14 SCREWS REQUIRED ON OPTIONAL PART 65-51617 COVER PAN).



VENDORS

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