

TO: ALL HOLDERS OF 737 POWER PLANT BUILDUP MANUAL, 71-03-01

REVISION NO. 65, DATED MAR 1/05

HIGHLIGHTS

DESCRIPTION OF CHANGE	TOPICS AFFECTED												
	D & O	D / A s s y	C l e a n i n g	I n s p / C h k	R e p a i r	A s s y	F / C	T e s t	T / S h o o t i n g	S / T o o l s	S t o r a g e	I P L	L / O v e r h a u l
<p>Changed v-band clamp installation instructions to reflect instructions in BAC 5001-9</p> <p>Changed torque values for clamp part numbers BACC10DU300AB and BACC10DP300A to reflect instruction in BAC 5001-9</p> <p>Added clamp part number BACC10DU300AE and torque value to Figure 606</p>						X							

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737 POWER PLANT BUILDUP

71-03-01

BOEING P/N 65-73769-1 thru -16, -19, -20, -23, -24, -29 thru -38,
 -41 thru -46, -51 thru -56, -61 thru -78

AIRLINE P/N

THE FOLLOWING DIRECTIVES APPLY TO THIS SUBJECT:

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		MC 3510-1	May 15/68
		MC 3913-18	May 15/68
		PRR 30186	May 15/68
		PRR 30401-1 Rev B	May 15/68
		PRR 30401-2 Rev B	May 15/68
		PRR 30662	May 15/68
		PRR 30681	May 15/68
		PRR 30717 Rev A	May 15/68
		PRR 30727 Rev A	May 15/68
78-1003		MC 3821-2	Aug 15/68
		MC 3821-4	Aug 15/68
		PRR 30606 Rev B	Aug 15/68
		PRR 30661 Rev B	Aug 15/68
36-1001		PRR 30926	Aug 15/68
75-1001		PRR 30934	Aug 15/68
		PRR 31090-1	Aug 15/68
		PRR 31129	Aug 15/68
		PRR 31135	Aug 15/68

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		PRR 30087	Feb 15/69
		PRR 30360	Feb 15/69
75-1001, Rev 1		PRR 30934, Rev C	Feb 15/69
		PRR 31030	Feb 15/69
		PRR 31030-12	Feb 15/69
75-1002		PRR 31129	Feb 15/69
		MC 3560-2	Aug 15/69
		MC 3580-2	Aug 15/69
		PRR 30087-1	Aug 15/69
		PRR 30571-2	Aug 15/69
		PRR 30606	Aug 15/69
		PRR 30727-1	Aug 15/69
		PRR 31090-1	Aug 15/69
		PRR 31135-1	Aug 15/69
29-1008		PRR 31484	Aug 15/69
71-1015		MC 3460-1K	Nov 10/69
		MC 3500-3	Nov 10/69
71-1011		PRR RC 10703	Nov 10/69
		PRR 31429-3	Nov 10/69
		PRR 31429-4	Nov 10/69
		PRR 31429-5	Nov 10/69
		PRR 31429-6	Nov 10/69
		PRR 31429-9	Nov 10/69
		PRR 31429-10	Nov 10/69
		PRR 31439	Nov 10/69
		PRR 31714	Nov 10/69
		PRR 31726	Nov 10/69
		PRR 31733	Nov 10/69
		PRR 31736	Nov 10/69
78-1005		PRR 31030K	Mar 10/70
76-1001		PRR 31079	Mar 10/70
		PRR 31429-5, Rev E	Mar 10/70
		PRR 31671	Mar 10/70
		PRR 31791	Mar 10/70
		PRR 31814	Mar 10/70
36-1004		PRR 31866	Jun 10/70
		PRR 31914	Jun 10/70
		MC 3550	Sep 10/70
		PRR 31030-59K	Sep 10/70
75-1003		PRR 31867	Dec 10/70
79-1002		PRR 31872	Dec 10/70
		PRR 31998	Jun 10/71
71-1022		PRR 32023	Jun 10/71
		PRR 32042	Jun 10/71
		PRR 32064-3	Jun 10/71
		MR 30094-2	Jun 10/71

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		PRR 31791-1	Sep 10/71
		PRR 32064-1	Sep 10/71
		PRR 32064-7	Sep 10/71
		PRR 32087	Sep 10/71
24-1020		PRR 31989	Dec 10/71
77-1001		PRR 31993	Dec 10/71
		RR 97029-1	Dec 10/71
		MC 3580-5	Mar 10/72
78-1005, Rev 1		PRR 31030K	Mar 10/72
71-1018		PRR 31814	Mar 10/72
21-1032		PRR 32087	Mar 10/72
		PRR 32106	Mar 10/72
		PRR 32106-1	Mar 10/72
		PRR 32109-2	Mar 10/72
		PRR 32144	Jun 10/72
		PRR 32167	Jun 10/72
73-1002		PRR 32175	Sep 10/72
		MC 3540-4	Dec 25/72
		PRR 32226	Mar 25/73
		MC 3510-14	Mar 25/73
		MC 3510-15	Mar 25/73
		MC 3821-8	Mar 25/73
		MC 3821-7	Jun 25/73
		PRR 32236	Jun 25/73
		PRR 32213	Sep 25/73
		PRR 32290-3	Sep 25/73
		PRR 32292	Sep 25/73
		MC 3510-16	Sep 25/73
		MC 3560-3	Sep 25/73
		PRR 32315	Dec 25/73
		PRR 32332	Mar 25/74
29-1024			Jun 25/74
		PRR 32354	Jun 25/74
		PRR 32356	Jun 25/74
		PRR 32381	Jun 25/74
		PRR 32285-6	Sep 25/74
26-1014		PRR 32369	Mar 25/75
		PRR 32370	Sep 25/74
		PRR 32383	Dec 25/74
		MC 3540-6	Jul 5/79
		PRR 32414	Jun 25/75
		PRR 32470	Jun 25/75
80-1002			Jun 25/75
71-1029		PRR 32201	Jun 25/75

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
29-1024, Rev 1			Dec 25/75
79-1003			Dec 25/75
71-1032		MC 3461-12K	Jul 5/79
80-1003		PRR 32449	Dec 25/75
		PRR 32408	Dec 25/75
		PRR 32551	Dec 25/75
		PRR 32383-1	Dec 25/75
		MC 3461-11K	Jul 5/76
71-1034		MC 3510-53K	Jul 5/76
26-1017		MC 3510-40K	Jul 5/76
71-1041		MC 3510-53K	Jan 5/77
71-1033, Rev 1		PRR 32414	Jan 5/79
71-1043			Jan 5/77
		PRR 32639	Jan 5/77
26-1017R1			Jul 5/77
		MC 3550-3	Jul 5/77
		PRR 32756-3	Jan 5/78
71-1050R1		MC 3510-32K	Jul 5/78
		MC 3510-65K	Jan 5/78
71-1048		MC 3510-69K	Jan 5/78
36-1007		MC 3510-35K	Jul 5/78
29-1024R2			Jul 5/78
		MC 3510-51	Jan 5/79
		MC 3510-72	Jan 5/79
		MC 3510-75	Jan 5/79
		MC 3510-76	Jan 5/79
71-1045		PRR 32731-1	Jan 5/79
		PRR 32756-10	Jan 5/79
		PRR 32756-9	Jan 5/79
		PRR 32827-3	Jan 5/79
		PRR 32832-5	Jan 5/79
		PRR 32860-1	Jan 5/79
30-1010, Rev 2		PRR 31849	Jan 5/79
		PRR 32771	Jan 5/79
		PRR 32768	Jan 5/79
		PRR 32775	Jan 5/79
		MC 3510-60	Jul 5/79
71-1053		MC 3510-14K	Jul 5/79
71-1055		MC 3510-51K	Jul 5/79
71-1058		MC 3510-70K	Jul 5/79
		MC 3510-82	Jul 5/79
		PRR 32756-13	Jul 5/79
71-1060		MC 3510-80K	Jan 5/80
		MC 3510-84K	Jan 5/80
		MC 3510-88	Jan 5/80

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
71-1060		MC 3510-92	Jan 5/80
77-1001			Jan 5/80
			Jan 5/80
75-1004		PRR 32825	Jan 5/80
		MC 3550-4	Jul 5/80
		MC 3540-6	Jul 5/80
		MC 3510-96	Jul 5/80
		MC 3510-98	Jul 5/80
		PRR 33041	Jan 5/81
		MC 3510-93	Jan 5/81
71-1044		MC 3510-65K	Jan 5/81
		MC 3510-104	Jan 5/81
		PRR 32950-25	Jan 5/81
		PRR 32950-40	Jan 5/81
		PRR 33049	Jan 5/81
		PRR 33140-5	Jul 5/81
		MC 3520-17	Jul 5/81
71-1070			Jul 5/81
		PRR 33175	Jan 5/82
		PRR 33050-4	Jan 5/82
		MC 3510-118	Jan 5/82
71-1064		MC 3461-7K	Jan 5/82
*[2]		MC 3461-21K	Dec 5/84
71-1078		MC 3510-124K	Jan 5/82
80-1003R1			Jan 5/82
SB 71-1069		PRR 33071	Jun 5/86
		PRR 33203	Jul 5/82
		MC 3550-5	Jul 5/82
71-1087		MC 3510-122K	Jul 5/82
		PRR 33163-1	Jul 5/82
71-1095		MC 3510-121K	Jan 5/83
80-1007			Jul 5/83
		PRR 33325	Jul 5/83
71-1082		MC 3510-149K	Jul 5/83
		PRR 33004-10	Dec 5/83
*[1]		MC 3540-11K	Dec 5/84
		MC 3540-12	Dec 5/83
71-1069		PRR 33071	Jun 5/86
71-1030			Dec 5/83
79-1004			Dec 5/83
		MC 3510-182K	Dec 5/83
36-1007R2			Mar 5/84
71-1114			Mar 5/84

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
71-1032		3461-12K	Dec 5/84
71-1055		3510-51K	Dec 5/84
71-1058		3510-70K	Dec 5/84
71-1064		3461-7K	Dec 5/84
71-1071		3461-26K	Dec 5/84
71-1092		3461-26K	Dec 5/84
71-1093		3461-28K	Dec 5/84
71-1100		3461-29K	Dec 5/84
71-1101		3461-30K	Dec 5/84
		PRR 33327	Dec 5/84
		PRR 33365	Dec 5/84
		MC 354-14K	Dec 5/84
		PRR 33163-2	Dec 5/84
		PRR 33519	Dec 5/84
	71-3		Mar 5/85
71-1125		MC 3461-28K	Jun 5/85
71-1115		MC 3461-28K	Jun 5/85
71-1093R1		MC 3461-28K	Jun 5/85
71-1102		MC 3500-9K	Jun 5/85
71-1111		MC 3500-9K	Jun 5/85
		MC 3500-10	Jun 5/85
71-1094		MC 3510-102K	Jun 5/85
71-1086		MC 3510-107K	Jun 5/85
		MC 3510-119	Jun 5/85
71-1135		MC 3510-121K	Jun 5/85
71-1080		MC 3510-121K	Jun 5/85
71-1088		MC 3510-123K	Jun 5/85
71-1079		MC 3510-130K	Jun 5/85
71-1097		MC 3510-137K	Jun 5/85
71-1098		MC 3510-141K	Jun 5/85
71-1081		MC 3510-125K	Jun 5/85
71-1104		MC 3510-124K	Jun 5/85
	71-4		Jun 5/85
71-1089		MC 3510-145K	Jun 5/85
		MC 3510-193K	Jun 5/85

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		MC 3510-183k	Jun 5/85
		MC 3510-146K	Jun 5/85
		MC 3510-199K	Jun 5/85
		MC 3510-144	Jun 5/85
		MC 3510-207K	Jun 5/85
		MC 3510-69K	Jun 5/85
		MC 3510-208K	Jun 5/85
		MC 3510-213K	Jun 5/85
		MC 3540-11K	Jun 5/85
		MC 3540-11K	Jun 5/85
		MC 3540-11K	Jun 5/85
		MC 3540-11K	Jun 5/85
		MC 3540-11K	Jun 5/85
		MC 3461-21K	Jun 5/85
		RR 97065-38	Jun 5/85
		RR 97031-24	Jun 5/85
		RR 97012-3	Jun 5/85
		MC 3510-144	Jun 5/85
		MC 3510-198K	Jun 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-11K	Sep 5/85
		MC 3540-15K	Sep 5/85
		PRR 33822	Sep 5/85
		MC 3461-28K	Mar 5/86
		MC 3461-28K	Mar 5/86
		MC 3461-44K	Mar 5/86
		MC 3461-21K	Mar 5/86
		MC 3461-38K	Mar 5/86
		MC 3510-211K	Mar 5/86
		MC 3550-6	Mar 5/86
		MC 3550-7	Mar 5/86
		MC 3510-219K	Jun 5/86
		MC 3510-126K	Jun 5/86
		MC 3510-161K	Jun 5/86
		MC 3510-146K	Jun 5/86
		MC 3510-212K	Jun 5/86
		MC 3510-217K	Jun 5/86
		MC 3510-223K	Jun 5/86
		MC 3510-229K	Jun 5/86

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
71-1099		MC 3510-153K	Jun 5/86
71-1096		MC 3510-149K	Jun 5/86
71-1119		MC 3510-201K	Jun 5/86
71-1108		MC 3510-176K	Jun 5/86
71-1107		MC 3510-156K	Jun 5/86
71-1137		MC 3530-15K	Jun 5/86
71-1140		MC 3461-45K	Jun 5/86
71-1154		MC 3510-225K	Jun 5/86
71-1101R1		MC 3461-30K	Jun 5/86
26-1025		MC 3510-171K	Jun 5/86
		MC 3510-197K	Jun 5/86
30-1029		PRR 33803	Jun 5/86
71-1131		MC 3461-38K & -39K	Jun 5/86
		PRR 33004-25	Jun 5/86
		PRR 33880-6	Jun 5/86
71-1151		MC 3461-28K	Jun 5/86
		PRR 33078	Jun 5/86
		MC 3530-12K	Jun 5/86
71-1159		MC 3530-217K	Sep 5/86
71-1161		MC 3510-230K	Sep 5/86
71-1163		MC 3510-228K	Sep 5/86
80-1047		MC 3540-11K	Sep 5/86
80-1050		MC 3540-11K & -14K	Sep 5/86
		MC 3510-196	Sep 5/86
		MC 3510-204	Sep 5/86
		MC 3510-221	Sep 5/86
		MC 3510-231K	Sep 5/86
		MC 3510-143	Sep 5/86
		MC 3510-165	Sep 5/86
		MC 3510-232K	Sep 5/86
71-1182		MC 3510-235K	Jun 5/87
		MC 3510-163	Sep 5/86
71-1173		MC 3510-222K	Dec 5/86
71-1174		MC 3510-226K	Mar 5/87
71-1162		MC 3510-85K	Dec 5/86
71-1180		MC 3510-236K	Dec 5/86
80-1045R1		PRR 33822	Dec 5/86
80-1051		MC 3540-11K	Mar 5/87
78-1041		PRR 34152	Mar 5/87

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
		RR 97126-2	Jun 5/87
		MC 7200MK3010	Jun 5/87
71-1192		MC 3510-239K	Sep 5/87
		MC 3461-28K	Jun 5/87
80-1048		MC 3461-21K	Jun 5/87
		MC 3540-11K	Jun 5/87
		MC 3530-18	Jun 5/87
80-1052		PRR 34315-5	Jun 5/87
		MC 3540-11K	Sep 5/87
		MC 7200MK3021	Sep 5/87
71-1179		MC 3461-28K	Sep 5/87
		PRR 34103-1	Sep 5/87
		PRR 34287	Dec 5/87
		PRR 34201	Jun 5/88
71-1210		MC 7200MK3039	Sep 5/88
		MC 7200MK3041	Jun 5/88
36-1013			Sep 5/88
71-1069R3		PRR 33071	Jun 1/95
		PRR 34504	Jun 1/95
	71-6		Dec 5/88
		MC 3510-213K	Mar 5/89
		MC 3510-215K	Mar 5/89
		MC 3510-218K	Mar 5/89
		MC 3510-220K	Mar 5/89
		MC 3510-223K	Mar 5/89
		MC 3510-224K	Mar 5/89
		MC 3510-227K	Mar 5/89
		MC 3510-228K	Mar 5/89
71-1189		MC 3510-172K	Jun 5/90
71-1219		MC 7200MK3071	Jun 5/90
71-1198		MC 5460MK3002	Jun 5/90
80-1055		MC 7700MK3005	Jun 5/90
71-1255		MC 7200MK3134	Jun 5/91
80-1057		MC 7700MK3005	Jun 5/91
71-1258		MC 7200MK3032	Sep 5/91
71-1263		MC 7200MK3158	Dec 5/91
71-1278		MC 5460MK3003	Jun 5/92
71-1289		PRR 35096R	
71-1312		MC 7200MK3051	Jun 1/95
26-1077		MC 2611MK3004	Mar 1/96
71-1337		MC 7200MK3158	Mar 1/96
71-1340		MC 7200MK3265	Mar 1/96
71-1357			Jun 1/97
71-1341			Jun 1/97
71-1351			Jun 1/97
71-1363		MC 7200MK3316	Mar 1/99
71-1369		MC 7200MK3284	Mar 1/99

*[1] SB'S 80-1004 THRU 80-1023, 80-1025 THRU 80-1029, 80-1032, 80-1034

*[2] SB'S 71-1063, 71-1065, 71-1067, 71-1083, 71-1105, 71-1128

BOEING SERVICE BULLETIN	BOEING TEMPORARY REVISION	OTHER DIRECTIVES	DATE DIRECTIVE INCORPORATED INTO TEXT
71-1391 737-71A1430		MC7200MK3031	Mar 1/01 Jul 1/02

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision
 F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01		116	Nov 15/68	508	BLANK
T-1	Jan 5/78	117	Nov 15/68	508A	Mar 5/88
T-2	Dec 10/71	118	BLANK	508B	Mar 5/88
T-3	Jul 5/79	201	Mar 5/87	508C	Mar 5/88
T-4	Jan 5/81	202	Aug 15/67	508D	Mar 10/70
T-5	Jun 5/86	203	Aug 15/67	509	Aug 15/67
T-6	Jun 5/85	204	May 15/68	510	Jan 5/83
T-7	Jun 5/86	205	Nov 1/00	510A	Dec 5/86
T-8	Jun 5/87	206	BLANK	510B	Mar 10/70
T-9	Mar 1/99	301	May 15/68	510C	Aug 15/67
T-10	Jul 1/02	302	Aug 15/69	510D	Mar 10/70
* LEP-1	Mar 1/05	303	Mar 5/84	511	Mar 1/01
* LEP-2	Mar 1/05	304	BLANK	512	Sep 5/88
* LEP-3	Mar 1/05	401	Jul 5/77	512A	Sep 5/88
* LEP-4	Mar 1/05	402	Jul 5/77	512B	Sep 5/88
* LEP-5	Mar 1/05	403	Mar 5/87	512C	Sep 5/88
* LEP-6	Mar 1/05	404	Mar 5/87	512D	BLANK
* LEP-7	Mar 1/05	405	Mar 5/87	513	Sep 5/88
* LEP-8	Mar 1/05	406	Mar 1/97	514	Sep 5/88
T/C-1	Jul 5/76	407	Mar 5/87	514A	Sep 5/88
T/C-2	BLANK	408	BLANK	514B	Sep 5/88
1	Jan 5/77	501	May 15/68	514C	Sep 5/88
2	May 15/68	502	Jun 5/85	514D	BLANK
3	Aug 15/67	502A	Mar 10/70	515	Aug 15/67
4	BLANK	502B	Mar 10/70	516	May 15/68
101	Nov 15/68	502C	Mar 10/70	516A	Dec 25/75
102	Sep 1/96	502D	Mar 10/70	516B	BLANK
103	Nov 10/69	* 503	Mar 1/05	516C	Nov 15/68
104	Dec 10/70	504	Jun 5/85	516D	Jul 5/82
104A	Jun 10/70	504A	Jul 5/81	517	Dec 5/84
104B	Aug 15/69	504B	Jul 5/81	518	Jul 5/82
105	Nov 15/68	504C	May 15/68	518A	Jul 5/82
106	Nov 15/68	504D	May 15/69	518B	BLANK
107	Dec 25/75	504E	Jul 5/81	518C	Aug 15/68
108	Nov 15/68	504F	BLANK	518D	Aug 15/68
109	Nov 15/68	504G	Mar 5/85	519	Dec 5/86
110	Nov 15/68	504H	Mar 5/85	520	Jul 5/78
111	Nov 15/68	505	Mar 5/85	520A	Jul 5/78
112	Nov 10/69	506	Mar 5/87	520B	Jul 5/78
112A	Nov 10/69	506A	Mar 5/87	520C	Jul 5/78
112B	BLANK	506B	Jun 5/88	520D	Jul 5/78
113	Nov 15/68	506C	Jun 5/88	520E	Jul 5/78
114	Nov 15/68	506D	Aug 15/67	520F	Jun 5/87
115	Nov 15/68	507	Mar 5/84	520G	Jul 5/78

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision
 F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		534C	Nov 10/69	543	Dec 5/86
520H	Mar 5/86	534D	May 15/68	544	BLANK
521	Sep 5/86	535	May 15/68	544A	Jun 5/85
522	Jun 10/70	536	Nov 10/69	544B	Mar 5/87
522A	BLANK	536A	Nov 10/69	544C	Mar 5/87
522B	Feb 15/69	536B	Jan 5/81	544D	Mar 5/87
522C	Jan 5/77	536C	Jan 5/81	544E	Mar 5/87
522D	BLANK	536D	Jan 5/81	544F	BLANK
522E	Sep 10/71	536E	BLANK	545	May 15/68
522F	Jul 5/76	536F	Jul 5/82	546	Aug 15/67
523	Dec 5/89	537	Nov 10/69	546A	Jan 5/80
524	Aug 15/67	538	Jan 5/81	546B	Mar 5/87
524A	Jun 10/72	538A	Jun 10/71	546C	Dec 5/93
524B	BLANK	538B	Jun 10/71	546D	Dec 5/93
524C	Aug 15/68	538C	Jul 5/81	547	Jun 5/85
524D	May 15/68	538D	Jul 5/81	548	Jun 1/97
525	Dec 10/70	538E	Jul 5/81	548A	Jun 5/85
526	BLANK	538F	BLANK	548B	Dec 5/93
526A	Jun 10/71	539	Jul 5/81	548C	Jun 5/86
526B	Jun 10/71	540	Jul 5/82	548D	Jun 5/86
526C	Jul 5/81	540A	Jan 5/81	548E	Jun 5/86
526D	Jan 5/81	540B	Dec 25/74	548F	Jun 5/86
527	Jan 5/81	540C	Jun 25/75	548G	Jun 5/86
528	Jul 5/82	540D	Jul 5/82	548H	Jun 5/86
528A	Jan 5/81	540E	Jul 5/82	548I	Jun 5/86
528B	Nov 10/69	540F	Jul 5/82	548J	Jun 5/86
528C	Dec 25/75	540G	Jan 5/81	549	Jun 5/86
528D	Jun 25/75	540H	Jun 1/95	550	Jun 5/86
529	Jan 5/81	540I	Jun 1/95	550A	Jun 5/86
530	Jul 5/82	540J	Nov 1/00	550B	Jun 5/86
530A	Jan 5/81	540K	Jun 1/95	550C	Jun 5/86
530B	BLANK	540L	Jun 1/95	550D	Jun 5/86
530C	Jun 10/71	540M	Jun 1/95	550E	Jun 5/86
530D	Mar 10/70	540N	Jun 1/95	550F	Jun 5/86
531	Jul 5/81	540O	Jun 1/95	550G	Jun 5/86
532	Jan 5/81	540P	Jun 1/95	550H	Jun 5/86
532A	Nov 10/69	541	Jun 1/95	551	Jun 5/86
532B	Nov 10/69	542	Jun 1/95	552	Jun 5/86
532C	Nov 10/69	542A	Jun 1/95	552A	Dec 5/87
532D	Jul 5/81	542B	Jun 1/95	552B	Dec 5/87
533	Jul 5/82	542C	Jun 1/95	552C	Dec 5/87
534	Jul 5/81	542D	Sep 1/96	552D	Dec 5/87
534A	Nov 10/69	542E	BLANK	552E	Jun 5/86
534B	Jun 25/75	542F	Jun 1/95	552F	Jun 5/86

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision

F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		562A	Feb 15/69	574	Aug 15/68
552G	Dec 5/87	562B	May 15/68	574A	Aug 15/67
552H	Dec 5/87	562C	Jun 5/87	574B	May 15/68
552I	Dec 5/87	562D	Jun 5/86	574C	Aug 15/67
552J	Dec 5/87	562E	Jun 5/86	574D	BLANK
552K	Dec 5/87	562F	BLANK	575	Mar 25/74
552L	Dec 5/87	563	Sep 5/85	576	May 15/68
552M	Dec 5/87	564	Jul 5/78	576A	Jul 5/81
552N	Dec 5/87	564A	Dec 5/86	576B	Mar 5/88
552O	Dec 5/93	564B	Jun 5/87	576C	Mar 5/88
552P	Dec 5/93	564C	Jul 1/04	576D	Mar 5/88
552Q	Dec 5/93	564D	Jun 5/85	576E	Mar 5/88
552R	Dec 5/87	564E	Jan 5/79	576F	Mar 5/88
552S	Dec 5/87	564F	Mar 5/91	576G	Mar 5/88
552T	BLANK	564G	Sep 10/71	576H	Mar 5/88
553	Jan 5/80	564H	Sep 10/71	577	Mar 5/88
554	Mar 5/84	564I	Mar 5/91	578	May 15/68
554A	Mar 1/96	564J	Jun 5/87	578A	May 15/68
554B	Jun 5/86	565	Jun 5/87	578B	Mar 5/84
554C	Jun 10/72	566	BLANK	578B1	Mar 5/84
554D	May 15/68	566A	Nov 1/00	578B2	Dec 5/84
555	Jun 5/86	566B	Aug 15/69	578B3	Mar 5/86
556	BLANK	566C	Jun 5/87	578B4	BLANK
556A	May 15/68	566D	Sep 5/87	578C	Mar 5/86
556B	May 15/68	567	Sep 5/87	578D	Jun 5/87
556C	Dec 5/86	568	BLANK	578E	Sep 5/89
556D	BLANK	568A	Mar 5/84	578F	Mar 5/86
557	Aug 15/67	568B	Jul 5/81	578G	Dec 5/86
558	May 15/68	568C	Jul 5/81	578H	Sep 5/85
558A	Jun 5/87	568D	May 15/68	579	Jun 5/86
558B	BLANK	569	May 15/68	580	Sep 5/88
558C	Dec 5/90	570	May 15/68	580A	Sep 5/88
558D	Sep 5/86	570A	Jun 5/87	580B	Sep 5/88
558E	BLANK	570B	Mar 5/85	580C	Jun 5/90
558F	Jun 5/91	570C	Jun 5/85	580D	Sep 5/88
559	Jun 5/91	570D	Dec 25/74	581	Sep 5/88
560	Dec 5/83	571	Jan 5/79	582	Jun 5/90
560A	Jun 5/91	572	May 15/68	582A	Jan 5/82
560B	Jun 5/91	572A	Sep 5/86	582B	Jan 5/82
560C	BLANK	572B	Mar 10/70	583	Sep 5/87
560D	Jan 5/81	572C	Jan 5/79	584	Jun 5/88
561	Jan 5/81	572D	May 15/68	584A	Jun 5/88
562	BLANK	573	Aug 15/68	584B	BLANK

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision

F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		596L1	Dec 5/86	604	Mar 10/70
584C	May 15/68	596L2	Jun 5/87	605	Mar 10/70
584D	May 15/68	596L3	Jun 5/87	606	Mar 10/70
585	May 15/68	596L4	Jun 5/87	* 607	Mar 1/05
586	Jun 5/85	596M	Jul 5/76	608	Jun 5/87
586A	Jun 5/85	596N	Mar 25/74	609	Jun 5/85
586B	Mar 5/84	596N1	Mar 25/74	610	BLANK
586C	Sep 5/87	596N2	Jun 5/87	701	Aug 15/67
586D	BLANK	596P	BLANK	702	BLANK
587	Jul 5/81	596Q	Mar 10/70	801	Aug 15/67
588	Jul 5/83	596R	Mar 10/70	802	BLANK
588A	Jul 5/83	596S	Mar 10/70	901	Mar 5/92
588B	BLANK	597	Jun 5/87	902	Mar 5/92
588C	Jun 5/86	598	Aug 15/69	1001	Dec 5/86
588D	Jun 5/86	598A	Feb 15/69	1002	Dec 5/86
589	Jun 5/87	598B	May 15/68	1101	Jan 5/83
590	Aug 15/67	598C	May 15/68	1102	Dec 5/86
590A	Jul 5/81	598D	Jun 1/95	1102A	Dec 5/86
590B	BLANK	598E	Jun 1/95	1102B	Mar 5/89
590C	Sep 5/86	598F	Dec 5/86	1102C	Jun 1/95
590D	Sep 5/86	598F1	Jun 5/87	1102D	Jun 1/95
591	Jun 5/87	598F2	BLANK	1102E	Mar 5/89
592	Dec 1/96	598G	Dec 5/86	1102F	Mar 5/89
593	Sep 5/86	598H	Jul 5/81	1102G	Jul 1/02
594	BLANK	598H1	Jun 5/87	1102H	Jun 1/95
594A	Mar 5/86	598H2	Jun 5/87	1103	Jun 1/95
594B	Jan 5/82	598H3	Jun 5/87	1104	Sep 5/84
595	Dec 5/86	598H4	Jun 5/87	1104A	Jun 5/92
596	Jun 5/85	598H5	Jun 5/87	1104B	Sep 5/86
596A	Jan 5/81	598H6	Jun 5/87	1104C	Dec 5/86
596B	Dec 5/84	598I	Mar 25/74	1104D	Sep 5/86
596C	Dec 5/84	598J	Mar 25/74	1104D1	BLANK
596D	BLANK	598K	Jun 5/87	1104D2	Dec 5/86
596E	Jun 5/87	598L	Mar 10/70	1104E	Sep 5/86
596F	Sep 1/97	598M	Feb 15/69	1104F	Jun 5/87
596G	Sep 1/97	598N	Feb 15/69	1104G	Jun 5/91
596H	Jun 1/95	598O	Feb 15/69	1104H	Sep 5/85
596I	Jun 1/95	598P	Mar 10/70	1105	Jun 5/86
596J	Dec 5/86	598Q	Mar 5/85	1106	Jul 5/81
596J1	Jun 5/87	598R	Mar 5/85	1106A	Dec 1/96
596J2	BLANK	601	Mar 10/70	1106B	Jul 5/81
596K	Dec 5/86	602	Mar 10/70	1106C	BLANK
596L	Jul 5/81	603	Jun 5/86	1106D	Mar 25/74

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision

F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		1114I	Sep 5/87	1124D	Jun 5/86
1107	Jul 5/81	1114J	Jan 5/81	1124E	Dec 5/87
1108	BLANK	1114K	Sep 5/87	1124F	Jun 5/86
1108A	Sep 10/71	1114L	Sep 5/87	1124G	Jun 5/86
1108B	Aug 15/68	1115	May 15/68	1124H	Jun 5/86
1108C	Jul 5/80	1116	May 15/68	1124I	Jun 5/86
1108D	Sep 5/86	1116A	Aug 15/69	1124J	Jun 5/86
1109	Dec 5/85	1116B	Aug 15/69	1124K	Dec 5/87
1110	Dec 5/85	1116C	Dec 5/86	1124L	Dec 1/95
1110A	Dec 5/85	1116D	Dec 5/86	1124M	Jun 5/86
1110B	Dec 5/85	1116E	Sep 5/91	1124N	Jun 5/86
1110C	Dec 5/85	1116F	BLANK	1125	Jun 5/86
1110D	Dec 5/85	1117	Jul 5/82	1126	Jun 5/86
1110E	Dec 5/85	1118	Feb 15/69	1127	Sep 5/87
1110F	Dec 5/85	1118A	Jul 5/82	1128	Dec 1/95
1110G	Dec 5/85	1118B	Mar 5/93	1128A	Dec 5/93
1110H	Dec 5/85	1118C	Sep 5/88	1128B	Jun 5/86
1110I	Dec 5/85	1118D	Mar 1/99	1128B1	Jun 5/86
1110J	BLANK	1118E	Mar 5/93	1128B2	Jun 5/86
1111	Dec 5/85	1118F	BLANK	1128B3	Jun 5/86
1112	Dec 5/85	1118G	Jun 10/72	1128B4	Jun 5/86
1112A	Nov 10/69	1118H	Jun 10/72	1128B5	Sep 5/87
1112B	Dec 5/85	1119	Jun 5/86	1128B6	Sep 5/87
1112C	Jun 1/97	1120	Jun 5/92	1128B7	Sep 5/87
1112D	Jun 1/97	1120A	BLANK	1128B8	Sep 5/87
1112E	Dec 5/85	1120B	Aug 15/69	1128B9	Dec 5/87
1112F	Sep 5/87	1120C	Jun 1/95	1128B10	Mar 1/99
1112G	Dec 5/85	1120D	Dec 5/86	1128C	Mar 1/99
1112H	Dec 5/90	1120E	Dec 5/86	1128D	Mar 5/86
1113	Dec 5/90	1120F	Dec 5/86	1128D1	Mar 5/86
1114	Dec 5/90	1121	Mar 5/88	1128D2	BLANK
1114A	Dec 5/85	1122	Mar 5/88	1128E	Sep 10/72
1114B	Dec 5/85	1122A	Mar 5/88	1128F	Feb 15/69
1114C	Jun 5/90	1122B	Mar 5/88	1129	May 15/68
1114D	Jun 5/90	1122C	BLANK	1130	Nov 15/68
1114E	Dec 5/85	1122D	Mar 5/88	1130A	Jun 5/91
1114F	Dec 5/86	1122E	Jun 5/92	1130B	Jun 5/91
1114F1	Dec 5/85	1122F	BLANK	1130C	Jun 5/91
1114F2	Dec 5/86	1123	Jun 5/86	1130D	Jun 5/92
1114F3	Jun 5/90	1124	Jun 5/86	1130E	BLANK
1114F4	Jun 5/90	1124A	Jun 5/86	1130F	Jun 5/87
1114G	Jun 1/97	1124B	Jun 5/86	1131	Jun 5/87
1114H	BLANK	1124C	Jun 5/86	1132	Jun 5/87

LIST OF EFFECTIVE PAGES

- * Indicates pages revised, added or deleted in latest revision
 F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		1140C	Jun 1/97	1150H	Jun 5/87
1132A	Jun 5/92	1140D	Jun 5/91	1150I	Jun 5/87
1132B	Jun 5/92	1140E	Sep 5/86	1150J	Jun 5/87
1132B1	BLANK	1140F	Sep 5/86	1150K	Mar 1/99
1132B2	Sep 5/87	1140G	Jun 5/90	1150L	Sep 1/97
1132C	Jun 1/95	1140H	Jun 5/87	1150M	Mar 1/99
1132D	Mar 5/87	1140I	Jun 5/87	1150N	BLANK
1132E	BLANK	1140J	Jun 5/92	1151	Sep 1/97
1132F	Jul 5/82	1141	Jan 5/80	1152	May 15/68
1133	Jun 5/88	1142	Mar 1/96	1152A	Sep 5/86
1134	May 15/68	1142A	Mar 1/96	1152B	Jul 5/81
1134A	Jun 10/70	1142B	Jan 5/80	1152C	Sep 5/86
1134B	Jan 5/77	1143	Mar 1/96	1152D	Jul 5/80
1134C	Dec 5/89	1144	Mar 1/96	1152E	Sep 5/86
1134D	Mar 5/91	1144A	Mar 1/96	1152F	Jun 1/95
1134E	Jun 5/87	1144B	BLANK	1152G	Jun 5/92
1134F	Sep 10/71	1145	Jul 5/80	1152H	Jun 1/95
1134G	Sep 10/71	1146	Jul 5/78	1152H1	Jun 1/95
1134H	Nov 10/69	1146A	Dec 5/90	1152H2	Jun 1/95
1135	Jun 5/90	1146B	Dec 5/90	1152I	Dec 5/85
1136	Jun 5/86	1146C	Mar 5/90	1152J	Dec 5/85
1136A	Jun 5/87	1146D	Sep 5/86	1153	Jul 5/81
1136B	Jun 5/90	1146E	Sep 5/86	1154	Jun 1/95
1136C	Jun 5/87	1146F	BLANK	1154A	Mar 5/89
1136D	Jun 5/87	1147	Sep 5/86	1154B	Mar 1/96
1137	Jun 5/87	1148	Jun 5/91	1154C	Mar 1/96
1138	Jun 5/87	1148A	Jun 5/91	1154D	Jun 1/95
1138A	Dec 5/84	1148B	Jun 5/91	1154E	Mar 1/96
1138B	May 15/68	1148C	Jul 5/81	1154F	BLANK
1138C	Jun 5/86	1148D	May 15/68	1155	Mar 5/93
1138D	Jan 5/83	1149	Mar 5/84	1156	Jun 1/95
1138E	Mar 10/72	1150	Jan 5/81	1156A	Mar 1/96
1138F	Mar 5/88	1150A	Sep 5/86	1156B	Dec 1/96
1138G	Mar 5/88	1150B	Sep 5/86	1156C	Jun 1/95
1138H	Jun 5/86	1150C	Dec 5/84	1156D	Jun 5/91
1139	Jun 5/86	1150D	Jun 5/87	1156E	Dec 5/87
1140	Jun 5/86	1150D1	Dec 5/84	1156F	Sep 5/87
1140A	Jun 1/97	1150D2	Jun 5/87	1156G	Sep 5/87
1140B	Mar 5/88	1150D3	Mar 5/87	1156H	BLANK
1140B1	Mar 5/88	1150D4	BLANK	1157	Jun 1/95
1140B2	Mar 5/88	1150E	Jun 5/87	1158	Jun 5/90
1140B3	Mar 5/88	1150F	Jun 5/86	1158A	Sep 5/86
1140B4	BLANK	1150G	Mar 1/99	1158B	Sep 5/86

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision

F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		1217	Dec 5/84	1240	Sep 5/86
1159	Sep 5/87	1218	Jun 5/86	1241	Jan 5/81
1160	Jun 1/97	1219	Jun 5/88	1242	Jun 1/95
1161	Jun 1/97	1220	Dec 5/83	1242A	Jul 5/83
1162	Mar 5/88	1220A	Jun 5/86	1242B	Jun 5/88
1163	Dec 5/93	1220B	Sep 5/86	1243	Jun 5/88
1164	Jun 1/95	1221	Jul 5/81	1244	Jun 5/88
1165	Dec 1/96	1222	Dec 5/84	1244A	Jun 5/86
1166	BLANK	1222A	Jun 5/87	1244B	Jun 5/85
1201	Jul 5/83	1222B	BLANK	1245	Dec 5/84
1202	Mar 5/91	1223	Jun 5/88	1246	Jan 5/81
1202A	Mar 5/91	1224	Jun 5/88	1247	Jan 5/81
1202B	Mar 5/91	1224A	Sep 5/87	1248	Mar 5/84
1203	Mar 5/91	1224B	BLANK	1248A	Mar 1/99
1204	Sep 5/84	1225	Sep 5/86	1248B	Jun 5/88
1205	Jul 5/83	1226	Jun 5/85	1249	Dec 5/83
1206	Jun 1/95	1226A	Jun 5/86	1250	Jun 5/86
1206A	Jul 5/83	1226B	BLANK	1250A	Dec 5/91
1206B	Jun 5/86	1227	Jun 1/95	1250B	BLANK
1207	Mar 1/99	1228	Jun 5/85	1251	Jun 5/86
1208	Mar 5/86	1228A	Mar 1/96	1252	Jun 5/86
1208A	Mar 5/89	1228B	Jun 1/95	1252A	Dec 5/84
1208B	Mar 5/89	1228C	Jun 1/95	1252B	BLANK
1208C	Mar 5/89	1228D	BLANK	1253	Dec 5/84
1208D	BLANK	1229	Jun 1/97	1254	Jul 5/81
1209	Jul 5/80	1230	Sep 5/86	1255	Jun 5/85
1210	Jul 5/80	1230A	BLANK	1256	Jun 5/86
1210A	Dec 5/84	1230B	Jun 5/90	1256A	Jun 1/95
1210B	BLANK	1231	Jun 5/87	1256B	Jun 5/85
1211	Jan 5/81	1232	Sep 5/86	1256C	Jun 5/87
1212	Dec 5/84	1233	Jun 1/95	1256D	BLANK
1212A	Mar 1/01	1234	Jun 5/85	1257	Jul 5/81
1212B	Jun 1/95	1234A	Dec 5/84	1258	Jan 5/81
1212C	Jun 5/85	1234B	Dec 5/84	1259	Sep 5/87
1212D	BLANK	1234C	Mar 5/89	1260	Sep 5/87
1213	Mar 5/85	1234D	Mar 5/89	1260A	Sep 5/87
1214	Jun 5/86	1235	Sep 5/91	1260B	BLANK
1214A	Jun 5/86	1236	Dec 5/84	1261	Sep 5/87
1214B	Jun 5/87	1237	Jun 5/85	1262	Dec 5/84
1215	Jun 5/88	1238	Jun 5/91	1262A	Dec 5/84
1216	Jun 5/87	1238A	Jun 5/91	1262B	Dec 5/86
1216A	Jun 1/95	1238B	Jun 5/87	1263	Jun 5/87
1216B	Jun 5/88	1239	Jun 1/97	1264	Dec 5/84

LIST OF EFFECTIVE PAGES

* Indicates pages revised, added or deleted in latest revision
 F Indicates foldout pages - print one side only

PAGE	DATE	PAGE	DATE	PAGE	DATE
71-03-01 (CONT)		1286D	BLANK		
1264A	Dec 5/86	1287	Jun 5/87		
1264B	Dec 5/84	1288	Sep 5/86		
1265	Jan 5/81	1289	Dec 5/84		
1266	Jun 5/86	1290	BLANK		
1267	Jan 5/81				
1268	Dec 5/83				
1269	Mar 5/89				
1270	Mar 5/89				
1270A	Mar 5/89				
1270B	Mar 5/89				
1271	Jan 5/81				
1272	Jun 5/88				
1272A	Sep 5/84				
1272B	Jun 5/88				
1272C	Jun 5/88				
1272D	Jun 1/95				
1272E	Jun 1/95				
1272F	BLANK				
1273	Sep 5/88				
1274	Dec 5/84				
1275	Jan 5/81				
1276	Jan 5/82				
1276A	Mar 5/89				
1276B	BLANK				
1277	Jun 5/85				
1278	Dec 5/84				
1278A	Jun 1/97				
1278B	BLANK				
1279	Mar 1/96				
1280	Jan 5/81				
1281	Jun 1/95				
1282	Jan 5/81				
1283	Dec 5/86				
1284	Dec 5/84				
1284A	Jun 5/84				
1284B	Mar 5/89				
1284C	Jun 1/95				
1284D	BLANK				
1285	Dec 5/84				
1286	Jun 5/85				
1286A	Jun 5/87				
1286B	Sep 5/88				
1286C	Sep 5/88				

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

TABLE OF CONTENTS

<u>Paragraph Title</u>	<u>Page</u>
Description and Operation	1
Disassembly	101
Cleaning.	201
Inspection/Check.	301
Repair.	401
Assembly.	501
Fits and Clearances	601
Testing	701
Trouble Shooting.	801
Storage Instructions.	901
Special Tools, Fixtures, and Equipment.	1001
Illustrated Parts	1101
QEC Index	1201

POWER PLANT BUILDUP

Boeing Part Numbers: 65-73769

DESCRIPTION AND OPERATION

1. General

- A. This subject provides a sequential procedure for complete buildup of the 737 JT8D-7, JT8D-9, JT8D-15 and JT8D-17 power plant. The procedure includes instructions for installation of all accessories contained in the quick engine change (QEC) kit pertinent to the engine in each position on the airplane, along with all items peculiar to specific engine configurations used by the different airlines.
- B. The subject is divided into sections in accordance with Specification ATA-100 as follows:
- (1) Disassembly - Complete instructions are given for dismantling each individual power plant (positions 1, port, and 2, starboard, on the airplane) to a maximum neutral configuration in par. 3 and 4. Paragraph 5 gives instructions for stripping the engine down from maximum neutral configuration to the basic engine.
 - (2) Cleaning - Gives detailed cleaning procedures. For a list of approved cleaning agents, refer to Chapter 20.
 - (3) Inspection/Check - Procedures are given for inspection of all accessories and components of the power plant to determine the necessity for repair or replacement.
 - (4) Repair - Provides complete repair and rework procedures required to return worn or damaged parts to serviceable condition.
 - (5) Assembly - Gives buildup procedures as follows:
 - (a) In paragraph 3, instructions (including illustrations) are given for preassembling many accessory components. These assemblies are made up of adjacent parts or components which logically go together to form an easily installed unit.
 - (b) Paragraph 5 gives instructions (including illustrations) for building up the basic engine to a maximum neutral configuration. The maximum neutral configuration is the basic engine plus those accessories and components common to both positions on the airplane. At this point the engine is interchangeable between both positions.
 - (c) In paragraphs 6 and 7, instructions are given for building up the engines from maximum neutral to final configurations for each position on the airplane. At this time, items peculiar to a particular airline are installed.

- (6) Fits and Clearances - Operator is referred to Pratt and Whitney JT8D Engine Overhaul Manual. Specific clearances are given throughout overhaul procedures where necessary.
- (7) Testing - Operator is referred to the Pratt and Whitney, JT8D Engine Overhaul Manual.
- (8) Troubleshooting - Operator is referred to the Pratt and Whitney, JT8D Engine Overhaul Manual.
- (9) Storage Instructions - Provides detailed instructions for depressuring and preparation for preserving, for both short and long periods of time.
- (10) Special Tools - Lists all special tools and equipment necessary for handling, disassembling, and assembling power plant.
- (11) Illustrated Parts List - Provides a complete list of all parts used to build up the engine, with exploded views of installation on engine.
- (12) A numerical parts list index is also provided on the end of this subject which lists all the parts in alpha-numerical order against the figure and index number where they are used.

2. Description

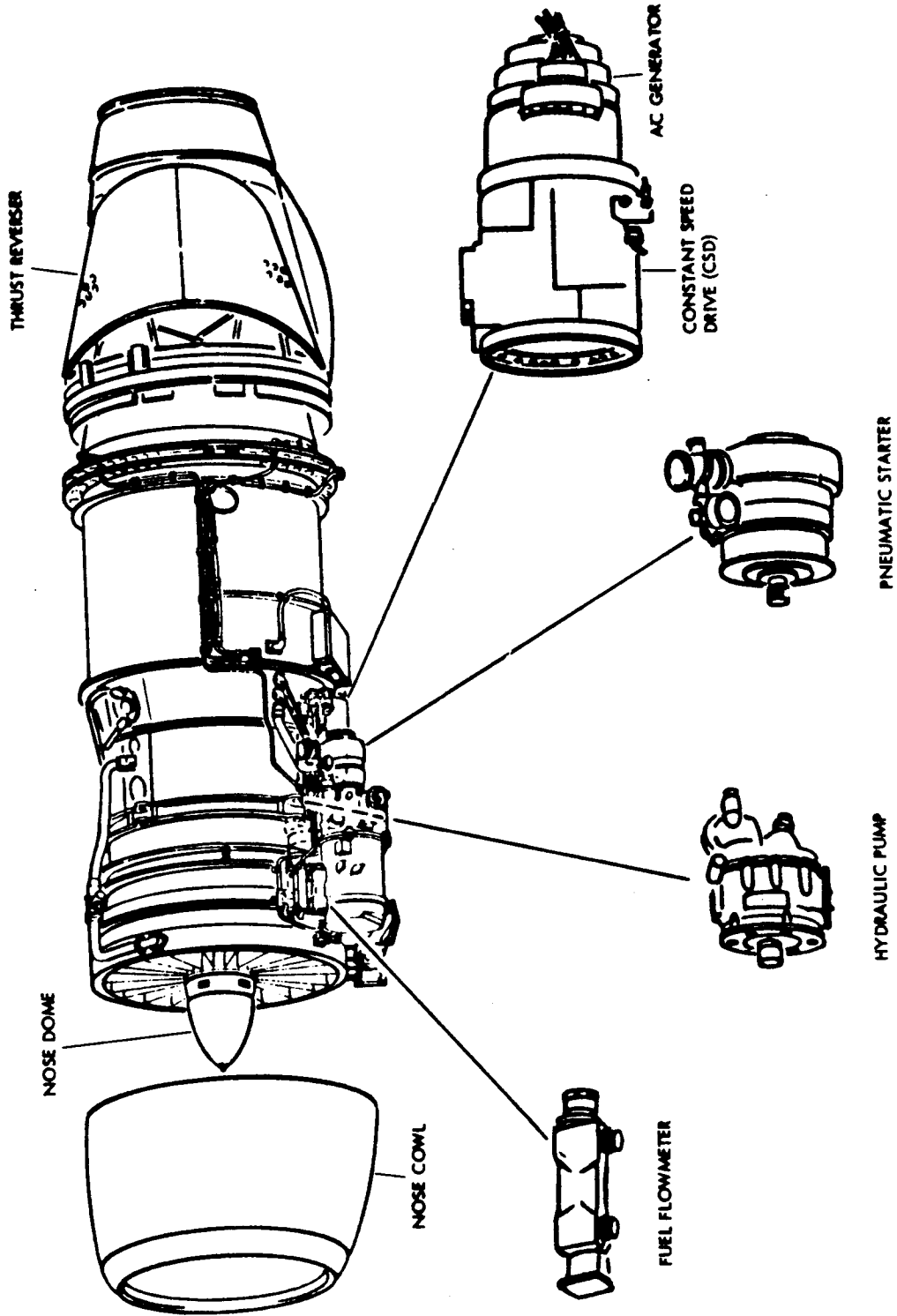
A. Leading particulars of the power plant and its major components are as follows:

(1) Weights

- (a) Bare basic engine -- 3057 lb
- (b) Thrust reverser -- 480 lb
- (c) Nose cowl -- 109 lb
- (d) Constant speed drive -- 87 lb
- (e) AC generator -- 87 lb
- (f) Starter -- 25 lb
- (g) Hydraulic pump and tubing -- 50 lb
- (h) Fuel flowmeter -- 20 lb
- (i) Tubing, hoses, wiring and miscellaneous -- 350 lb
- (j) Engine complete - 4265 lb

(2) Dimensions

- (a) Built-up engine length -- 17.8 ft
- (b) Maximum diameter of built-up engine -- 49.5 in
- (c) Maximum radial projection from engine centerline -- 35.0 in



Power Plant - Major Breakdown
Figure 1

Aug 15/67

DISASSEMBLY

1. General

A. Instructions are given in paragraphs 3 and 4 for removing components peculiar to specific engines (positions 1 and 2 on the airplane). In paragraph 5 procedures are laid out for stripping the engine down from this configuration to the bare, basic engine by removing all accessories and components of engine common to either position on the airplane. The general course of disassembly, after removal of nose cowl and thrust reverser, is left side, rear to front; right side, front to rear; bottom; then support brackets, both Pratt and Whitney and Boeing, common to the Boeing engine installation. Paragraph 6 gives instructions for removing brackets.

B. General Disassembly Practices

- (1) Install covers, plugs, or caps on all engine openings, tubes, hoses, and electrical connectors immediately upon removal of each accessory or component.
- (2) Ensure that every part removed from the engine is correctly identified. If part number inscribed on part is not visible, or readily identifiable, attach an identification tag to the part.
- (3) It is not necessary to remove brackets installed by the engine manufacturer, but it is recommended that they be removed for proper inspection, and to ensure proper replacement when necessary.

2. Preparation for Disassembly

A. Ensure that all necessary tools, equipment, and handling fixtures are available. Refer to "Special Tools," page 1001.

B. Provide covers for the following engine openings:

- (1) N1 Tachometer Generator Pad
- (2) Fuel Flowmeter Adapter
- (3) Oil Tank Sight Gage Pads
- (4) Oil Quantity Transmitter Pad
- (5) Oil Tank Pressure-Filling Fitting Pad

- (6) Gearbox Overboard Vent Pad
- (7) Starter Pad
- (8) Constant-Speed Drive (CSD) Pad
- (9) Hydraulic Pump Pad
- (10) N2 Tachometer Generator Pad
- (11) Fuel Heater Exhaust Mounting Pad
- (12) Generator Cooling Air Pad
- (13) TAI and Air Conditioning Manifold Pads
- (14) Fuel Inlet Pad
- (15) Engine Inlet Opening
- (16) Aft Turbine Flange

C. Provide caps and covers for all electrical connectors, tubes, and hoses.

D. Place engine in a suitable handling fixture.

E. Drain oil tank.

F. Drain constant-speed drive case. Reinstall drain plug.

G. Remove the cavity standpipe cap on the engine gearbox and remove the wet spline cavity drain plug. Drain the oil from the CSD spline cavity. Reinstall the cavity standpipe cap finger tight. Reinstall the wet spline cavity drain plug and tighten to 190-210 lb-in.

H. Drain accessory drive gearbox.

3. Removal of Components From Engine No. 1 Only

A. Remove nose cowl as follows: (See figure 562.)

CAUTION: NOSE COWL ASSEMBLY WEIGHS APPROXIMATELY 110 POUNDS. SUPPORT ENTIRE WEIGHT OF THE ASSEMBLY WITH A SUITABLE SLING OR DOLLY WHILE REMOVING FASTENERS.

(1) Disconnect TAI and vortex dissipator ducts.

(2) Remove bolts (1 and 5).

(3) Remove nose cowl (2) and store in suitable holding fixture or dolly.

B. Remove potable water pressure plumbing as follows: (See figure 565.)

- (1) Remove union (6) from upper end of tube assembly (3).
- (2) Disconnect tube assembly (3) from fitting in TAI duct just above thermostatic valve. Remove screws (5), clamps (4), and tube assembly (3).
- (3) Remove special fitting (1) and gasket (2) from port in TAI duct.

C. Remove nose cowl thermal anti-icing system components as follows: (See figure 565, alternate configuration.)

- (1) Disconnect tube assembly (22) at union (18) and elbow (12). Remove screws (24), clamps (23) and tube assembly (22).
- (2) Remove hose assembly (21).
- (3) Remove nut (20), washer (19), and bulkhead union (18) from bracket on flange "D."
- (4) Remove elbow (15), nut (17), and washer (16) from valve (13).
- (5) Remove valve (13) and gasket (14) from duct on left side of engine, just forward of flange "F."
- (6) Remove elbow (12) from forward port of filter and valve assembly (7).
- (7) Disconnect TAI hose from top of filter and valve assembly (7).
- (8) Disconnect hose fitting from port in TAI duct just above thermostatic valve, and remove screws (8), nuts (9), and filter and valve assembly (7).

NOTE: Disassemble filter and valve assembly (7) on bench. See figure 517 for details of assembly.

4. Removal of Components From Engine No. 2 Only

A. Remove nose cowl as follows: (See figure 567.)

CAUTION: NOSE COWL ASSEMBLY WEIGHS APPROXIMATELY 110 POUNDS. SUPPORT ENTIRE WEIGHT OF THE ASSEMBLY WITH A SUITABLE SLING OR DOLLY WHILE REMOVING FASTENERS.

- (1) Disconnect TAI and vortex dissipator ducts.
- (2) Remove bolts (1 and 5).

(3) Remove nose cowl (2) and store in suitable holding fixture or dolly.

B. Remove plug from TAI duct as follows: (See figure 570.)

(1) Remove plug (1) and gasket (2) from port in TAI duct.

5. Removal of accessories and components from basic engine.

A. Removal of thrust reverser.

| (1) Deleted.

(2) Remove hydraulically operated thrust reverser as follows: (See figures 563 and 568.)

(a) Disconnect hydraulic lines, push-pull cable and wire bundle at thrust reverser.

(b) Remove brackets (6) by removing bolts (7), washers (8) and nuts (9).

- (c) Remove thrust reverser (1) by removing bolts (2), washers (3 and 4) and nuts (5). Place thrust reverser in transportation dolly F80101 or equivalent.

CAUTION: THE THRUST REVERSER WEIGHS APPROXIMATELY 386 POUNDS. SUPPORT ENTIRE WEIGHT OF UNIT DURING REMOVAL BY USING HANDLING SLING F80100 OR EQUIVALENT.

NOTE: Refer to Subject 78-30-12 for overhaul instructions for hydraulically operated thrust reverser.

- B. Remove fire detector and wire bundle from tailpipe. (See figures 564 and 569.)

- (1) Disconnect wire bundle (1 or 7) from fire detector element and from engine wire bundle.
- (2) Remove fire detector element from clamps.

NOTE: Handle element carefully. Store in a safe place.

- (3) Remove wire bundle (1 or 7), clamps (2 and 3 or 8 and 9), screws (4 or 10) and nuts (5 or 11).

- C. Remove tailpipe assembly and shroud assembly. (See figure 558.)

CAUTION: WEAR CLEAN, WHITE, COTTON GLOVES WHEN HANDLING TITANIUM PARTS. ANY SALT DEPOSIT, EVEN FINGERPRINTS, ON UNPREPARED SURFACE CAN CAUSE TITANIUM ALLOY TO CORRODE WHEN SUBJECTED TO HIGH TEMPERATURES.

- (1) Remove tailpipe assembly (1) and shroud assembly (10) from engine by removing bolts (12).

- D. Remove tail plug (1, figure 556) by removing bolts (2).

- E. Remove hydraulic tubes as follows: (See figure 555.)

- (1) Remove clamps (5) and spacers (6 and 7) by removing screws (8 and 9).
- (2) Remove tube assembly (4).
- (3) Remove nut (3), washer (2), and elbow (1).

- F. Remove turbine vibration pickup as follows: (See figure 554.)

- (1) Disconnect electrical plug D0374 from vibration pickup (1).
- (2) Remove screws (3), and remove vibration pickup. If adapter (2) is used, remove with vibration pickup. Cut lockwire and detach adapter from pickup.

- G. Remove aft engine drain fitting assemblies as follows: (See figure 551.)
- (1) Disconnect fire detector element from wire bundle.
 - (2) Remove clamps (22) and screws (23).
 - (3) Remove screws (18) and nuts (19), and remove either bracket (17) or clamp assembly (21).
 - (4) Remove bracket assembly (13) by removing nuts (16), washers (15), and screws (14).
 - (5) Detach wire bundle from drain fitting bracket by removing nut (12), screw (11), and clamp (10).
 - (6) Disconnect hose (10, figure 553) from drain fitting assembly (6).
 - (7) Remove drain fitting assembly (6) by removing nuts (9), washers (8), and screws (7).
 - (8) Disconnect tube assembly (5, figure 553) from drain fitting assembly (1).
 - (9) Remove drain fitting assembly (1) by removing nuts (4), washers (3), and screws (2).

H. Remove CSD oil cooler as follows: (See figure 550.)

- (1) Detach hoses of oil cooler assembly (1) from elbows (5 and 8) in bracket on flange "K."
- (2) Remove nuts (4) and washers (3). Remove oil cooler assembly (1) and gasket (2).

NOTE: Oil cooler assembly (1) will be disassembled on bench. See figure 504 for detail parts of assembly.

- (3) Detach tube assembly from elbow (5). Remove elbow (5) by removing jamnut (6) and washer (7).
- (4) Detach tube assembly from elbow (8). Remove elbow (8) by removing jamnut (9) and washer (10).

I. Remove P & D drain tube and combustion chamber fuel drain tubes as follows: (See figure 553.)

- (1) Remove clamps (11, 12, 13, and 14) and spacer (15) by removing nuts (18) and screws (16 and 17).
- (2) Remove hose (10).
- (3) Remove elbow (7) and jamnut (6) from combustion chamber fuel drain port FDI. Remove O-Ring (8) and backup ring (9).
- (4) Remove tube assembly (5).
- (5) Remove elbow (2) and jamnut (1) from P & D drain port. Remove O-Ring (3) and backup ring (4).

J. Remove CSD oil pressure system plumbing as follows: (See figure 552.)

- (1) Remove tube assembly (7).
- (2) Remove tube assembly (6).
- (3) Remove hose (5).
- (4) Remove hose (4).
- (5) Remove nut (3), washer (2), and union (1) from bracket on CSD oil filter support.

K. Remove CSD oil filter assembly as follows: (See figure 549.)

- (1) Detach ring of dust cap chain from filter support bracket.
- (2) Remove nuts (5), washers (4), and bolts (2). Remove spacers (3) and oil filter assembly (1).
- (3) Scrape off Metal-Cal (6) with a sharp-edged plastic or wooden tool.

NOTE: Disassemble oil filter assembly (1) on bench. See figure 503 for details of assembly.

L. Remove engine control linkage cranks as follows: (See figure 548.)

- (1) Remove nuts (11), washers (10), and two clamp halves (9).3
- (2) through (4) deleted.
- (5) Cut lockwire and remove nut which holds engine start crank (2) in place. Remove engine start crank, and reinstall nut.
- (6) Cut lockwire and remove nut which holds engine power crank (1) in place. Remove engine power crank and reinstall nut.

M. Remove cabin air and engine starter ducts as follows: (See figure 547.)

- (1) Remove hose (39).
- (2) Disconnect electrical plug D0402 from valve on manifold assembly (29). Remove bonding jumper (40) by removing screws (41), washers (42) and nuts (43).
- (3) Detach manifold assembly (29) from duct assembly (14) by removing clamp (38).
- (4) Detach manifold assembly (29) from bracket on flange "G" by removing nuts (37), washers (36), and bolts (35). Remove manifold assembly.
- (5) Remove clips (30 and 31) from manifold assembly (29) by removing screws (32), spacers (33), and washers (34).

NOTE: Disassemble manifold assembly on bench. See figure 509 for details of assembly.

- (6) Cut lockwire and loosen bearing nut at joint between manifold assembly (1) and duct assembly (14).
- (7) Cut lockwire and loosen bearing nut at joint between duct assembly (14) and engine starter.

- (8) Remove nut (28), washer (27), bolt (26), and spacer (25).
- (9) Remove clamp (24) and remove duct assembly (14).
- (10) Remove rod end bearings (11 and 12) from duct assembly (14) by removing nut (17), washer (16), and bolt (15). Remove jamnut (13) and separate rod end bearings (11) and (12).
- (11) Remove nuts (10), washers (9), bolts (8) and spacers (7).
- (12) Cut lockwire and remove bolts (2, 3, and 4) and washers (5 and 6). Remove manifold assembly (1).

NOTE: Disassemble manifold assembly (1) on bench. See Fig. 508 for details of assembly.

N. Remove engine starter as follows (Fig. 546):

- (1) Detach electrical plugs D0424 and D0422 from starter. Disconnect coupling (7) and remove retainer (4) with shutoff valve (5). Remove containment assembly (18).
- (2) Remove nuts (3).
- (3) Remove starter (1) and gasket (2).

O. Remove engine oil pressure differential switch as follows (Fig. 545):

- (1) Detach electrical plug D0754 from pressure switch (5).
- (2) Detach tube assemblies of pressure switch assembly from fitting in ports LP5 and LP66.
- (3) Remove screws (6), and remove pressure switch assembly (5).

NOTE: Disassemble pressure switch assembly (5) on bench. See Fig. 512 for details of assembly.

- (4) Remove universal bolts (2), universal elbows (1), and O-rings (3 and 4) from ports LP5 and LP6.

P. Remove engine oil temperature transmitter as follows (Fig. 544):

- (1) Detach electrical plug D0726 from temperature transmitter (1).
- (2) Cut lockwire and remove temperature transmitter (1). Remove O-ring (2).

- Q. Remove oil pressure transmitter and switch assembly as follows: (See figure 543.)
- (1) Detach electrical plug D0684 from transmitter.
 - (2) Detach electrical plug D0752 from switch.
 - (3) Remove nuts (17 and 22), screws (16 and 21), and clamps (15, 18, 19, and 20).
 - (4) Remove tube assembly (14).
 - (5) Remove elbow (2), jamnut (1), O-ring (3), and backup ring (4) from gearbox transmitter oil vent port LV3.
 - (6) Detach hose from union (5) in oil pressure line low pressure warning port LP3.
 - (7) Remove union (5) and O-ring (6).
 - (8) Remove transmitter and switch assembly (7) by removing nuts (10 and 13), washers (9 and 12), and screws (8 and 11).
- NOTE: Disassemble transmitter and switch assembly (7) on bench. See figure 505 for details of assembly.
- R. Remove inlet vibration pickup as follows: (See figure 542.)
- (1) Detach electrical plug D0376 from vibration pickup (1).
 - (2) Remove screws (3). Remove vibration pickup (1). If adapter (2) is used, remove adapter also. Cut lockwire and detach adapter from pickup.
- S. Remove gearbox overboard vent fitting (2, figure 541) and gasket (1) by removing nuts (4) and washers (3).

T. Remove accessory components from engine oil tank as follows: (See figure 540.)

- (1) Remove nuts (41), washers (40), and bolts (39).
- (2) Remove pressure filling fitting assembly made up of O-rings (33 and 36), fitting (34), quick-disconnect coupling half (35), and dust cap (37). Detach clip (38) and disassemble all components of assembly.
- (3) Remove nuts (32), washers (31), and bolts (30).
- (4) Remove pressure filling fitting assembly made up of O-rings (24 and 27), fitting (25), quick-disconnect coupling half (26), and dust cap (28). Detach clip (29) and disassemble all components of assembly.
- (5) Scrape off metal-cals (42 and 43) with a sharp-edged wooden or plastic tool.
- (6) Remove nuts (23), washers (22), and bolts (21).
- (7) Remove and disassemble sight glass units, each made up of cap (20), glass (19), reflector (18), fitting (16), washers (17), and O-ring (15).
- (8) Detach electrical plug D0682 from tank unit (1) in bottom of tank.
- (9) Remove screws (14), and remove clamps (13) which attach tank unit wire bundle.

NOTE: Place a waste container beneath oil tank to catch residual fluid which drains when tank unit is removed.
- (10) Remove nuts (10), bolts (5, 8, 9, and 12), and washers (6 and 7).
- (11) Remove bonding jumper (4), bracket (3), and brackets (11). Slide tank unit (1) out of tank.

U. Remove fuel flowmeter as follows: (See figure 539.)

- (1) Detach electrical connectors.
- (2) Detach bonding jumper (17) from flowmeter attaching plate on flange "B" by removing nut (20), washer (19), and screw (18).
- (3) Remove bonding jumper (17) by removing nut (26) and screw (25).
NOTE: If bonding jumper (24) is used, remove by removing nut (28) and screw (27).
- (4) Remove aft support bracket (1) installed by engine manufacturer.
NOTE: Place a waste container beneath flowmeter to catch residual fluid which drains when flowmeter is removed.
- (5) Remove four bolts which attach lockplate (2) on forward end of flowmeter.
- (6) Remove entire flowmeter assembly and disassemble on bench.
- (7) Install P & W flowmeter adapter in place of flowmeter.

V. Remove engine drain plumbing as follows (See figure 538.)

- (1) Disconnect tube assembly (21) from elbow (17) in gearbox overboard drain port LD5. Remove elbow (17), nut (18), O-ring (19), and backup ring (20).
- (2) Disconnect tube assembly (24) from union (22) in oil cup overflow drain port LD3. Remove union (22) and O-ring (23).
- (3) Remove screws (26), and clamps (25).
- (4) Remove tube assembly (16).
- (5) Remove tube assembly (15).
- (6) Remove tube assembly (14).
- (7) Remove tee (10), nut (11), packing (12), and backup ring (13) from accessory drive overboard drain port LD6.
- (8) Remove tube assembly (9).
- (9) Remove tee (5), nut (6), O-ring (7), and backup ring (8) from fuel pump drain port LD6.
- (10) Remove drain manifold (1) by removing nuts (4), washers (3), and screws (2).

- W. Remove fuel heater air exhaust duct as follows: (See figure 537.)
- (1) Disconnect fire detector sensor from wire bundle.
 - (2) If bracket (13) has been installed, remove by removing screws (14) and nuts (15). Remove jumper assembly (16).
 - (3) Remove either bracket (7), or clamp assembly (12), by removing screws (8) and nuts (9).
 - (4) Remove seal (6).
 - (5) Remove nuts (5) and washers (4). Remove bracket assembly (3), duct assembly (2), and gasket (1).
- X. Cut lockwire and remove rigging pin mount (1, figure 536) by removing bolts (2 and 3) and washers (4).
- Y. Remove fuel inlet tube as follows: (See figure 535.)
- (1) Remove screws (8), and clamps (7).
 - (2) Detach tube assembly (2) from bracket on flange "C" by removing nut (6).
 - (3) Cut lockwire and remove nuts (5). Remove washers (4), flange (1), tube assembly (2), and gasket (3).

Z. Remove nose cowl thermal anti-icing system and vortex dissipator plumbing as follows: (See figure 534.)

(1) From engines without vortex dissipator:

- (a) Detach electrical plug D0196 from connector on thermostatic valve.
- (b) Remove clamps (8) by removing nuts (10) and screws (9).
- (c) Remove clamp (5).
- (d) Remove clamp (6), and remove duct assembly (4).
- (e) Remove duct assembly (7). Remove ring (11) from duct assembly.
- (f) Remove nuts (3) and remove clamp (2).
- (g) Remove two bolts attaching thermostatic valve assembly (1) to bracket on flange "D." Remove assembly.

NOTE: Disassemble thermostatic valve assembly (1) on bench. See figure 518 for details of assembly.

(2) From engines with vortex dissipator:

- (a) Detach electrical plugs: D0196 from connector on thermostatic valve (12), D1544 from connector on vortex dissipator control valve (39), and D1546 from connector on pressure switch (56 or 64).
- (b) Remove tube assembly (62 or 66) from tee (52) and union (50).
- (c) Remove pressure switch (64), if used, and O-ring (65).
- (d) Remove tube assembly (61).
- (e) Remove clamp (63), pressure switch (56), clamp (57), and screw (58).
- (f) Remove coupling (44), gasket (43), clamp (45), screw (46), washer (47), nut (48), and duct assembly (42). Remove ring (49) from duct.
- (g) From duct assembly (42), remove union (59), O-ring (60), tee (52), O-ring (53), and nut (54).
- (h) Remove cap (55) from tee.
- (i) Remove coupling (41), gasket (40), and valve (39).

- (j) Remove union (50) and O-ring (51) from valve.
- (k) Remove bolts (27), washers (28), and nuts (29) to release duct assembly (16) at thermostatic valve.
- (l) Remove clamp (30), screw (31), washer (32), and nut (33).
- (m) Remove coupling (25).
- (n) Remove clamp (20), clamp (21), spacer (22), screw (23), and nut (24).
- (o) Remove two clamps (17), four screws (18), four nuts (19), and duct assembly (16).
- (p) Remove gasket (26) from thermostatic valve.
- (q) Remove clamps (35), screws (36), nuts (37), and TAI duct assembly (34). Remove ring (38) from duct.
- (r) Remove two bolts (13), nuts (15), and washers (14) attaching thermostatic valve to bracket on flange "D," and remove valve (12). Install bolts, washers, and nuts loosely in flange of valve.

NOTE: Disassemble thermostatic valve assembly (12) on bench. See figure 518 for details of assembly.

AA. Remove air conditioning sensing plumbing as follows: (See figure 533.)

- (1) Remove screws (20 and 21), nuts (22), and clamps (18 and 19).
- (2) Remove tube assembly (4).
- (3) Remove nut (3), washer (2), and union (1).
- (4) Remove tube assembly (8).
- (5) Remove nut (7), washer (6), and elbow (5).
- (6) Remove hose (9).
- (7) Remove hose (16).
- (8) Remove nuts (12 and 15), washers (11 and 14), and elbows (10 and 13).
- (9) Remove tube assembly (17).

AB. Remove strut drain tubing as follows: (See figure 532.)

- (1) Remove clamps (23 and 26) and spacer (27) by removing screws (24 and 28), and nuts (25).
- (2) Remove union (22).
- (3) Remove tube assembly (21).
- (4) Remove union (20).
- (5) Remove tube assembly (19).
- (6) Remove union (18).
- (7) Remove tube assembly (17).
- (8) Remove hose (13).
- (9) Remove clamps (14) by removing nuts (16) and screws (15).
- (10) Remove tube assembly (12).
- (11) Remove nut (11), washer (10), and union (9).
- (12) Remove nut (6), washer (5), and tee (4).
- (13) Remove tube assembly (7).
- (14) Remove hose (8).
- (15) Remove nut (3), washer (2), and elbow (1).

AC. Remove N2 tachometer generator as follows: (See figure 531.)

- (1) Detach electrical plug D0552 from tachometer generator (2).
- (2) Remove nuts (4) and washers (3). Remove tachometer generator (2) and gasket (1).

AD. Remove hydraulic system components as follows: (See figure 530.)

- (1) Detach electrical plug D0766 from hydraulic pump.
- (2) Remove clamps (24, 25, and 26) and spacer (27) by removing nuts (30) and screws (28 and 29).
- (3) Detach all hoses from hydraulic pump assembly (1).
- (4) Remove pump assembly (1) as follows:

NOTE: If pump P/N 10-60470-7 is used perform steps (a) and (b) only. If any other pump is used, perform step (c) only.

- (a) Remove coupling (7). Remove hydraulic pump.
 - (b) Remove adapter (6) by removing nuts (3) and washers (4). Remove packing (5).
 - (c) Remove nuts (3) and washers (4). Remove hydraulic pump (1) and gasket (2).
- (5) Remove hose (18).
 - (6) Remove hose (17).
 - (7) Remove nut (16) and coupling halves (14 and 15).
 - (8) Remove nut (13) and coupling halves (11 and 12).
 - (9) Remove nut (10) and coupling halves (8 and 9).
 - (10) Remove filter assembly (19) by removing nuts (22), washers (21), and bolts (20).

NOTE: Disassemble filter assembly (19) on bench. See figure 516 for details of assembly.

AE. Remove differential pressure regulator as follows: (See figure 529.)

- (1) Remove clamps (9 and 10) and spacer (11) by removing nuts (13) and screws (12).
- (2) Remove hose (8).
- (3) Detach tube assembly from tee (5). Remove tee (5), nut (7), and washer (6).
- (4) Remove pressure differential regulator assembly (1) by removing nuts (4), washers (3), and bolts (2).

NOTE: Disassemble pressure differential regulator assembly (1) on bench. See figure 513 for details of assembly.

POWERPLANT BUILDUP

AF. Remove 8th stage air conditioning manifolds as follows: (See figure 528.)

- (1) Cut lockwire and remove bolts (13) and washers (14). Remove manifold assembly (12).

NOTE: Disassemble manifold assembly (12) on bench. See figure 507 for details of assembly.

- (2) Remove bolts (9), washers (10), and shim (8).
- (3) Remove manifold assembly (1) and bracket assembly (2) by removing bolts (3, 4, and 5) and washers (6 and 7).

NOTE: Disassemble manifold assembly (1) on bench. See figure 506 for details of assembly.

AG. Remove air conditioning precooler assembly as follows: (See figure 527.)

- (1) Remove bonding jumper (18) by removing nut (20), washer (19), nut (26), and screw (25).
- (2) Remove clamps (23), and remove flexible ducts (22) and duct (21).
- (3) Remove adapter (17) and seal (16) by removing remainder of nuts (20) and washers (19).
- (4) Remove nuts (15) and bolts (13 and 14). Remove support angle (12).
- (5) Remove bonding jumper (8) by removing nuts (11), screws (9), and washers (10).
- (6) Remove precooler assembly (1) by removing nuts (4 and 7) and bolts (2, 3, 5, and 6).

NOTE: Disassemble precooler assembly (1) on bench. See figure 510 for details of assembly.

AH. Remove fire detector assembly as follows (See figure 526.)

- (1) Disconnect sensor element assembly from electrical wire bundle at each end. Remove attaching hardware at each end.
- (2) Remove sensor assembly from quick-release clamps.

AI. Remove pressure ratio plumbing as follows: (See figure 524.)

- (1) Remove nuts (25), screws (24), and clamps (23).
- (2) Remove tube assembly (22).
- (3) Remove nut (21), washer (20), and union (19).
- (4) Remove tube assembly (18).
- (5) Remove cap (17) from tee (14).
- (6) Detach tube assembly (8) from tee (14). Remove nut (15), washer (16), and tee (14).
- (7) Remove tube assembly (8).
- (8) Remove nut (5), washer (6), and union (4).
- (9) Detach tube (7) from union (1). Remove nut (2), washer (3), and union (1).
- (10) Remove tube assembly (7).
- (11) Remove cap (13) from universal elbow (9).
- (12) Cut lockwire and remove universal bolt (10), universal elbow (9) and O-rings (11 and 12).

AJ. Remove ac generator cooling ducts as follows: (See figure 523.)

- (1) Detach bonding jumper (9) by removing nut (24), washer (23), and bolt (22).
- (2) Remove clamps (21), and remove semiflexible ducts (19 and 20) and duct (18).
- (3) Remove nuts (17), washers (16), and bolts (15).
- (4) Remove nuts (14), washers (12 and 13), and bolts (10 and 11). Remove bonding jumper (9) and bracket (8).
- (5) Pull cooling air duct (7) out of alternator air cooling port, and remove O-ring (6).
- (6) Remove bracket assembly (2) by removing nuts (5), bolts (3), and washers (4).
- (7) Loosen T-bolt and trunnion halves on collector ring assembly (1). Slide collector ring back and off rear end of generator.

NOTE: Disassemble collector ring assembly (1) on bench. See figure 502 for details of assembly.

AK. Remove ac generator and constant-speed drive assembly as follows: (See figure 522.)

CAUTION: SUPPORT ENTIRE WEIGHT OF GENERATOR AND CSD ASSEMBLY WHILE REMOVING ATTACHMENTS.

- (1) Loosen lug bolt (10) on QAD ring. Slide generator and CSD assembly (9) away from engine. Remove all parts of QAD assembly.

NOTE: Disassemble ac generator and CSD assembly (9) on bench. See figure 501 for details of assembly.

AL. Remove cone bolts as follows: (See figure 521.)

- (1) Remove aft cone bolt (1) by removing nut (5), bolt (2), and washers (3 and 4).
- (2) Remove forward cone bolts (7) by removing nuts (11), bolts (8), and washers (9 and 10).

AM. Remove nose dome as follows: (See figure 561.)

- (1) Loosen four captive nuts in nose dome assembly, and remove nose dome.
- (2) Remove PT2 coupler (1) and O-rings (2).

AN. Remove N1 tachometer generator as follows: (See figure 560.)

- (1) Detach electrical plug D0556 from connector on tachometer generator (2).
- (2) Remove nuts (4) and washers (3), and remove tachometer generator.

AO. Remove wire bundles and electrical fittings as follows: (See figure 525.)

- (1) Remove all supporting clamps and disconnect all plugs.
- (2) Disconnect all grounds from engine.
- (3) Remove wire bundles from engine.
- (4) Remove heat shield (6), bracket (2) and current transformer (1) by removing attaching hardware.

AP. Remove all brackets by removing attaching hardware. Reinstall all flange bolts removed to detach brackets. See figure 520 for all parts.

CLEANING

1. General

- A. Before cleaning, engine must be inspected for any damage, evidence of fluid leaks, missing parts, and other defective conditions, and such conditions recorded. Refer to "Inspection/Check."
- B. After removing accessories, plumbing, and wiring, clean exterior for storage preparation or shipping to engine overhaul.
- C. Thoroughly clean QEC kit items and other components removed from engine prior to their inspection and storage for future use in a power plant assembly.
- D. Refer to steam cleaning in 20-30-03 and vapor degreasing in 20-30-01 for cleaning information.

2. Cleaning Before Disassembly

- A. Fit plugs or caps over engine openings as follows:
 - (1) Cover engine inlet and engine exhaust.
 - (2) Tie covers over fuel heater air outlet and accessory gearbox breather outlet.
 - (3) Tie cover over AC generator cooling air outlet duct.
 - (4) Fit caps over all open hydraulic, fuel, and pneumatic tubes. Cover all engine openings, including starter outlet, open engine bleed pads, and CSD oil cooler. Wrap all loose electrical plugs in plastic sheet, and secure so that no cleaning fluid or foreign material can enter open ends of plugs. Secure plugs to engine to prevent damage during cleaning.
- B. Remove large areas of dirt by hand, with rags, or with a nonmetallic brush.
- C. Wash engine down with petroleum solvent spray, Specification P-D-680 (or equivalent). If no floor drain is provided for recovery of solvent, provide solvent catch trays.

WARNING: DO NOT USE EXCESSIVE SPRAY PRESSURE OR EXCESSIVE QUANTITIES OF FLUID.

CAUTION: OBSERVE ALL NECESSARY PRECAUTIONS WHEN SPRAYING FLAMMABLE SOLVENT. PROVIDE SUITABLE PROTECTIVE CLOTHING FOR PERSONNEL AND ENSURE ABUNDANT VENTILATION.

- D. Dry with air blast and rags. Remove plugs and caps installed before cleaning.
- E. Clean floor if power plant is to be disassembled at this same location.

3. Clean Basic Engine

- A. Fit covers over engine inlet and engine exhaust openings. Check that all other engine openings were plugged or sealed during disassembly. Seal or plug openings as necessary.
- B. Clean engine with a petroleum solvent spray, Specification P-D-680 (or equivalent), or use a low pressure steam jet.

CAUTION: DO NOT OPERATE CLEANING EQUIPMENT SO THAT SOLVENT OR STEAM IS FORCED INTO ENGINE OPENINGS.

NOTE: If additive compound is used in steam generator, rinse engine with clean water after cleaning.

- C. Dry engine with a hot air blast.

CAUTION: AIR VELOCITY MUST NOT BE GREAT ENOUGH TO ROTATE ENGINE.

- D. Remove covers and plugs from engine.

4. Clean Plumbing and Accessories

- A. Wipe the following items clean with rags and install protective shipping plugs. Send items to be overhauled in accordance with vendor's instructions.

- (1) Engine starter
- (2) Tachometer generators
- (3) Fire detector (Handle with extreme care, do not kink, bend or attempt to straighten.)
- (4) Hydraulic pump
- (5) Low pressure switch and oil filter bypass switch
- (6) Oil pressure transmitter
- (7) AC generator

- (8) Constant speed drive unit
- (9) Oil quantity tank unit
- (10) Oil and fuel temperature bulbs
- (11) Vibration pickups
- (12) All electrically operated valves, pressure regulators, and check valves

- B. Steam clean or vapor degrease all brackets and reusable hardware. Remove fluid drain fittings and steam clean or vapor degrease. Follow steam cleaning with a clean water rinse. After vapor degreasing, rinse with clean Turco 4215 alkaline cleaner or equivalent. Rinse in clean hot water above 160°F. Dry all parts in a warm air blast.

WARNING: TRICHLORETHYLENE IS TOXIC AND A MILD ANESTHETIC. AVOID PROLONGED OR REPEATED CONTACT WITH THE SKIN, OR BREATHING SOLVENT VAPOR. WEAR FULL FACE RESPIRATORY EQUIPMENT AND PLASTIC OR RUBBER GLOVES AND APRONS WHEN MANUALLY FLUSHING PARTS EITHER EXTERNALLY OR INTERNALLY WITH LIQUID SOLVENT.

CAUTION: VAPOR DEGREASING OF TITANIUM ALLOY PARTS IS PROHIBITED FOR PARTS WHICH WILL BE SUBJECTED TO TEMPERATURES OF 600°F OR ABOVE DURING SUBSEQUENT PROCESSING (i.e. STRESS RELIEVING, ANNEALING OR WELDING).

WHEN NONMETALLIC SPACERS OR HOLDERS ARE REQUIRED IN THE DEGREASER, USE NYLON POLYETHYLENE OR OTHER MATERIAL THAT WILL NOT SOFTEN OR DECOMPOSE IN TRICHLORETHYLENE.

WATER IN THE DEGREASER CAUSES FAULTY OPERATION OF DEGREASER AND CAUSES TRICHLORETHYLENE TO DECOMPOSE. DO NOT PROCESS WET PARTS, OR ALLOW WATER TO CONDENSE ON THE COOLING COILS.

- C. Clean ferrous brackets and clamps, which have been attacked by rust with a soft grit blast, then coat with a corrosion preventative substance.
- D. Clean metal tubes and fittings as follows:
- (1) Vapor degrease all metal tubes and fittings.
 - (2) Flush exterior and interior of tubes with clean liquid solvent.
 - (3) Blow out solvent remaining in each tube with dry, filtered air.

- (4) Flush metal tubes internally with filtered Turco 4215 alkaline cleaner, or equivalent. Flush with clean hot water, above 160°F.

NOTE: Ultrasonic cleaning in a continuously filtered bath of Turco 4215, followed with a hot water rinse, is an optional method. During ultrasonic cleaning, parts must be exposed to cavitating solution only. Use solution of 0.75 to 2.25 ounces of Turco 4215 alkaline cleaner per gallon of water, at temperature between 145°F and 190°F for a minimum of three minutes.

- (5) After rinsing, dry tubes and fittings with dry, warm, filtered air.

E. Clean flexible tubes and hoses as follows:

- (1) Flush interior of flexible hoses with a degreasing compound, Apco Thinner, or equivalent, for a maximum of ten minutes. Blow out solvent with dry, filtered air.
- (2) Flush flexible tubes internally with filtered Turco 4215 alkaline cleaner or equivalent for a maximum of ten minutes.
- (3) Dry flexible tubes with dry, warm, filtered air.

F. Immediately after cleaning and drying, cover all openings in part with dry, annealed aluminum foil. Seal and secure foil to the part with a suitable hot-dip coating. Other procedures providing equal protection may be used at discretion of operator.

5. Clean Constant Speed Drive Oil Cooler

- A. Clean and flush CSD oil cooler with kerosene in normal direction of flow, then in reverse direction before inspection.

6. Clean PT2 tube assembly as follows:

- A. Wipe outside with lint-free cloth and dry cleaning solvent. Do not immerse in liquid.
- B. Pass warm, dry, low pressure air (less than 10 psi) through the tube to assure that it is dry and clean inside.
- C. Clean O-ring groove receptacle with lint-free cloth and dry cleaning solvent.

- I 7. Clean engine electrical harness using TT-N-95, Type 1 solvent.
 - A. If, by visual examination, it is determined that a connector/receptacle interface has excessive contamination and connector replacement is not necessary use the following procedure:
 - (1) Materials
 - (a) Freon TF degreaser -- MS-180 (used to remove minor contamination of grease and oil) *[1]
 - (b) Freon T-P35 solvent -- MS-160 (used to remove major contamination of grease and oil and other contaminants) *[1]
 - (c) Extension hose, nozzle, solvent spray brush -- MS-226 *[1]
 - (d) Silicone lubricant -- Dow Corning No. 4 compound *[2]

*[1] Miller Stephenson Chemical Co., Inc.
George Washington Hwy, P.O. Box 950
Danbury, Connecticut 06810

*[2] Dow Corning Corp.
3902 S. Saginaw Road, P.O. Box 997
Midland, Michigan 48640

WARNING: FREON TF IS NON-FLAMMABLE AND NON-TOXIC, BUT SHOULD BE USED WITH ADEQUATE VENTILATION AND PROLONGED BREATHING SHOULD BE AVOIDED.

FREON T-P35 IS NON-TOXIC, BUT IS FLAMMABLE AND EXPLOSION MAY OCCUR. IT MUST NOT BE USED WHEN FLAMMABLE CONDITIONS EXIST. USE WITH ADEQUATE VENTILATION AND PROLONGED BREATHING SHOULD BE AVOIDED.

BE CAREFUL WHEN ASSEMBLING HOSE ASSEMBLY ON AEROSOL CAN TO AVOID GETTING SOLVENT INTO EYES.

- (1) Use brush assembly to apply solvent and brush the connector face until contaminants have been dissolved.
- (2) Flush connector with solvent to assure it is clean.
- (3) Allow connector to dry for one hour to allow solvent to dissipate. Tip connector to drain out solvent.
- (4) To ease connector/receptacle mating lubricate inner O-ring with Dow Corning No. 4 lubricant compound.

INSPECTION/CHECK

1. Check Before Power Plant Disassembly

A. Perform preliminary visual inspection/check of the engine before cleaning or disassembling. Note damage, unusual or excessive leaks and any other unsatisfactory conditions. Analyze damage areas, and endeavor to determine cause of damage or defect. Record results of inspection/check on a suitable form.

- (1) Check engine and attached components for damage, misalignment and distortion, loose attaching parts and broken bolts, and chafing or abrasion of metal parts and flexible hoses.
- (2) Check for excessive heat damage and air leaks. Check for loose connections between external air lines, and blown gaskets.
- (3) Check lubrication system for leaks, loose connections or damaged oil lines. Inspect oil tank for leaks. Examine oil filters for metal particles.
- (4) Check fuel system for leaks, loose connections or damaged fuel lines. Inspect fuel filter for evidence of contamination.
- (5) Check magnetic plug of constant speed drive unit sump for contamination.
- (6) Check hydraulic system for leaks and loose connections.
- (7) Check electrical harnesses for damage, chafed areas, loose wires and oil impregnation.
- (8) Check CSD oil filter for metal particles.
- (9) Check turbomag plugs for metal particles.

2. Examination of Engine Before Buildup
 - A. Before installation of any QEC kit components, inspect the basic engine as received from manufacturer or overhaul shop, and make a complete report of any missing plumbing items, bent brackets, or otherwise damaged parts.
 - B. Record the general external condition of the engine. Take particular note that storage seals and protective covers are unbroken and the desiccant indicator indicates a dry atmospheric condition inside the engine.
3. Standard Inspection/Check of Components Used for Powerplant Buildup (Quick Engine Change Kit)
 - A. Inspect all QEC assemblies, subassemblies, bracket assemblies, brackets, tubing, ducting, electrical wiring, electrical connectors, plugs, terminals, flexible hose assemblies and all supporting, mounting and attaching parts for the following conditions.
 - (1) Corrosion/discoloration
 - (2) Cracks/fractures
 - (3) Distortion/misalignment
 - (4) Deterioration/wear
 - (5) Separation/fatigue
 - (6) Security/looseness
 - (7) Heat damage/leaks
 - (8) Legibility of placards or missing placards
 - (9) Electrical continuity/insulation damage
 - (10) Damaged threads
 - B. During an inspection/check, note minor damage, such as scratches, small dents, or scores to brackets or on the surfaces of drain and vent tube assemblies. Do not designate plumbing parts subjected to pressurization for REPAIR, if the surface damage is extensive, or if the repair necessitates substantial reduction of tube wall thickness. (See figure 407.)

4. Inspection/Check After Disassembly
 - A. Examine all brackets for cracks and distortion.
 - B. Examine all attaching hardware for wear and damage.
 - C. Examine all sheet metal parts for scratches, dents, cracks, and corrosion. Use a strong light and 10-power magnification.
 - D. Fluorescent dye check all fittings.
 - E. Examine all tube assemblies for scratches, dents, and corrosion. Examine end fittings and nuts for stripped or crossed threads. See Fig. 407 for allowable damage.
 - F. Examine all ducts for scratches, dents, and corrosion. See Fig. 407 for allowable damage.
 - G. Discard all O-rings, gaskets, and packings.
 - H. Examine all wire bundles for breaks in insulation, broken shielding, broken wires, and broken terminals.
 - I. Examine all threaded parts for stripped or crossed threads.
 - J. Refer to manufacturer specifications for inspection of all valves, switches, electrical accessories, CSD, AC generator, starter, pumps and cone bolts.
 - K. Examine PT2 tube inlet for damage and test for leakage with 10 psi air pressure.

REPAIR AND REPLACEMENT

1. Repair

A. Repair of Steel Brackets

(1) Reshape brackets as follows:

CAUTION: DO NOT CLAMP BRACKET IN A SHARP-JAWED VISE. USE ONLY A SOFT FACED Mallet OR HAMMER TO BEND OR STRAIGHTEN FLANGES.

- (a) Determine correct angle and bend radius of all bends in bracket from engineering drawing, or from a new part if available.
- (b) Straighten all flanges and web of bracket by flattening on a smooth surface with a mallet.
- (c) Reform all straight bends with a mechanical bend break. Use correct bend radius.
- (d) To reform all other bends, and straight bends if a mechanical bend break is not available, clamp part in a vise and bend with a mallet.

NOTE: Deleted.

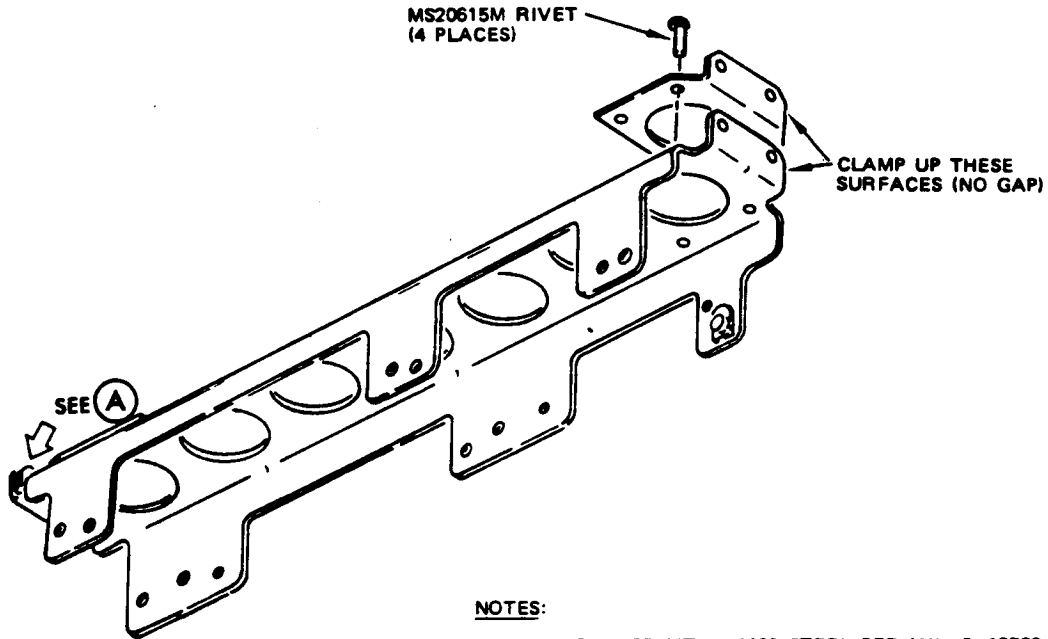
(2) Inspect bend areas to ensure no cracks exist.

- (a) See Fig. 400 for repair of 69-52261-1 bracket.

(3) Remove minor corrosion by polishing with No. 400 grit emery cloth or crocus cloth.

(4) Remove corrosion that emery cloth will not remove as follows:

- (a) Coat affected area with Oakite 31, Kelite Process K, Greater Mountain Chemical #801, or Turco Prepaint, diluted 1 part to 3 parts of water. Allow coat to remain on part several minutes or until coat turns a gray color.
- (b) Wipe part clean with a damp cloth and dry with clean lint-free cloth.

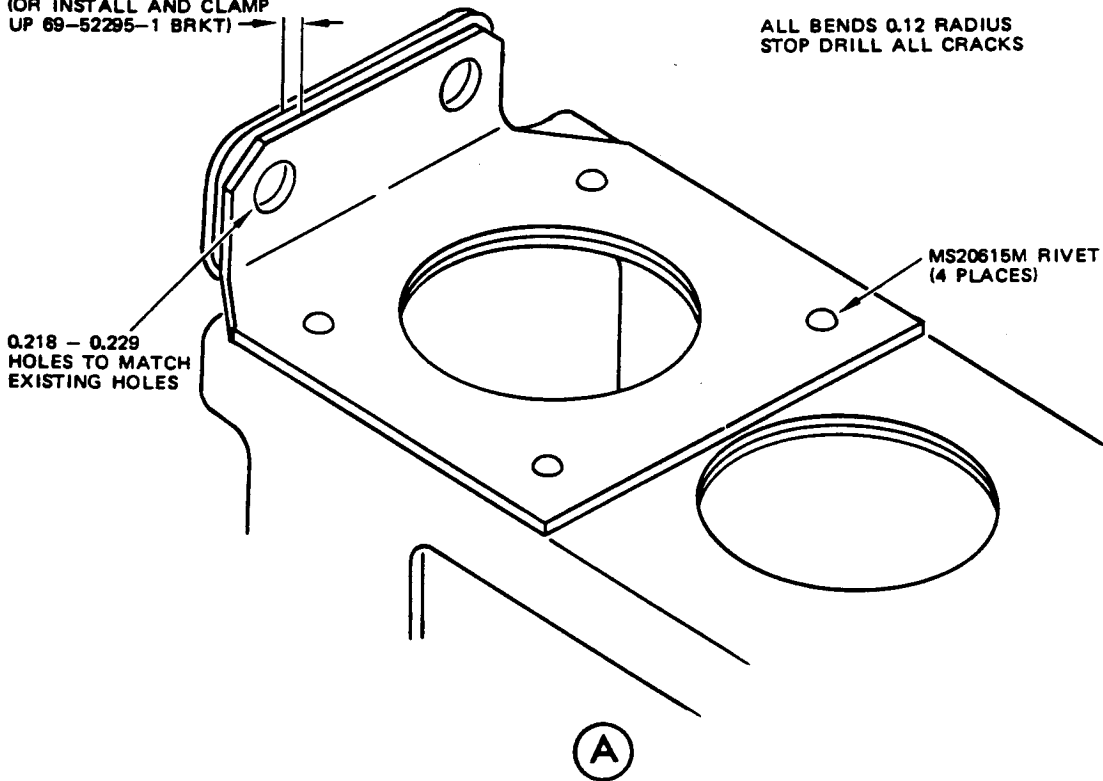


NOTES:

REPAIR DOUBLER MTL: 4130 STEEL PER MIL-S-18729. CADMIUM PLATE (0.0003 INCH THICK) WITH POST-PLATE CHROMATE TREATMENT PER 20-42-05), AND BAKE 3 HOURS MINIMUM AT 375 (± 25)°F. APPLY ONE COAT OF BMS 10-11, TYPE 1 PRIMER PER 20-41-02.

0.060 - 0.052 GAP
(OR INSTALL AND CLAMP
UP 69-52295-1 BRKT)

ALL BENDS 0.12 RADIUS
STOP DRILL ALL CRACKS



Bracket Repair Details
 Figure 400

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

- (c) Brush cadmium plate per 20-42-05, Method 3 or apply one coat of primer and two coats of aluminized lacquer. Allow a drying period of 45 minutes between each coat of primer and lacquer.

NOTE: EC-843 may be used in place of lacquer. 3M company, St. Paul, Minnesota. Apply as directed in Boeing Specification BAC 5797.

- (5) Remove surface pits or scratches and burrs within allowable limits by polishing with No. 240 grit emery cloth. After removing defect, polish area with No. 400 grit emery cloth or crocus cloth.

NOTE: See Fig. 401 for table of allowable defects.

- (6) Clean part after rework with solvent, BMS 3-2. Wipe part dry with clean, lint-free cloth.
- (7) Rubber stamp part number on bracket.
- (8) Coat bracket with corrosion preventive compound, MIL-C-11796, Class 3.
- (9) Wrap part in nonabsorbent material and identify with tag.

Burrs ▷	Rework if burr exceeds 0.015 inch on materials greater than 0.040 inch thick	Rework if burr exceeds 0.010 inch on materials less than 0.040 inch thick
Surface Pits ▷	Replace part if total surface area of individual pit exceeds 0.0014 square inch	Replace part if depth exceeds 0.05 inch or 5% of base metal thickness
Inclusions Having No Sharp Points ▷	Replace part if section thickness is reduced by 15% or more	Replace part if defect is greater than 0.010 inch in any dimension
Porosity	Replace parts if total area of porosity within any 1 x 1 inch square is greater than 0.02 square inch	Replace part if single area of porosity is greater than 0.01 square inch

▷ There must be at least 0.25 inch separation between edges of adjacent defects.

Bracket Damage Limits
 Figure 401

B. Repair of Aluminum Alloy Brackets

- (1) Reshape brackets as directed in par. A, step (1).
- (2) Inspect bend areas to ensure no cracks exist. Refer to INSPECTION/CHECK section.
- (3) Remove corrosion from brackets as follows:
 - (a) Polish affected area with aircraft polish, Specification MIL-P-6888.

NOTE: In lieu of polish, Kelite Process K, Oakite 36, or Turco W.O., No. 1 may be used per 20-30-02.
 - (b) Mix five to six parts of cleaning solvent, Specification BMS 3-2, Type I, or P-D-680, with 1 part emulsion cleaner per 20-30-02. Use mixture to clean polish from corrosion pits. Rinse part thoroughly with clear water.
 - (c) Inhibit further corrosion by applying a water solution containing 8 oz per gallon of potassium dichromate and 1/4 oz per gallon of Nacconal 40F or 90F. Allow solution to dry by evaporation. Do not wipe or rinse. Brush off excess potassium dichromate crystals.
- (4) Remove surface pits or scratches and burrs within allowable limits by polishing with No. 400 grit emery cloth. After removing defect, polish area with crocus cloth.

NOTE: See Fig. 401 for table of allowable defects.
- (5) After rework, clean part with solvent, Specification BMS 3-2. Wipe part dry with clean, lint-free cloth.
- (6) Rubber stamp part number on bracket.
- (7) Coat bracket lightly with corrosion preventive compound, Specification MIL-C-11796, Class 3.
- (8) Wrap part in nonabsorbent material and identify with tag.

C. Repair of Ducts and Tubes

NOTE: Thin-walled ducts and tubes are subject to damage from improper handling or abnormal operating conditions. In some cases, complete replacement of components is more desirable than repair or partial replacement.

- (1) No repair is required for slight surface scratches, pits, etc., or for smooth dents not exceeding allowable flattening.

NOTE: See Fig. 407 for allowable damage.

- (2) Repair parts per 36-10-03.
- (3) Machined flanges and formed sheet flanges may be reworked provided all dimensions, tolerances, and finish requirements of the drawing are met. Formed sheet metal flanges may be reworked provided minimum gage after rework is 0.022 inch or greater.
- (4) Drain tubes with dents and bends that do not seriously restrict flow are acceptable. Check reshaped drain tubes against new part or drawing. Open ends of tube should be reshaped if bent, and filed to remove burrs.
- (5) Flexible hoses used for drains may be reworked to extent of replacing damaged ends. There is no recommended repair for damaged high-pressure flexible hoses.
- (6) Upon completion of repair or rework of tubing and ducting, identify part as it appeared initially and code by appropriate color-coded tape per 20-50-10. Reclean tubing in accordance with subsection 4.D. on cleaning metal tubing and place in storage for future use.

- D. Repair sheet metal parts according to standard repair procedures established in Boeing 737 Structural Repair Manual, Chapter 51-40.
 - E. Repair accessories and purchased items according to manufacturer's specifications, or return to manufacturer for repair.
 - F. Repair minor damage to PT2 nose dome tube assembly as shown in Repair of Tubes. Repair leaks by fusion welding. Penetrant inspect all repair welds per Subject 20-20-02.
 - G. Repair wire bundles as required in accordance with procedures in Boeing Document D6-54446.
2. Refinish
- A. If plated or painted surfaces are worn or chipped but damage does not extend into base metal, refinish parts per applicable drawing finish specification.
 - B. Restore high temperature coating on damaged areas of titanium tailpipe extension per instructions in Subject 78-38-43.
3. Replacement
- A. Replace all parts worn or damaged beyond simple repair.
 - B. Replace all O-rings and rubber or composition gaskets and seals at each overhaul.
 - C. Replace all self-locking nuts.
 - D. Replace all dated hoses.
 - E. Replace all cotter pins at each overhaul.
 - F. Replace all parts worn beyond allowable limits.
 - G. Replace all tubes and ducts damaged beyond limits set forth in Fig. 407.

Figures 402 through 406 DELETED

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

SYSTEMS OPERATING PRESSURE	TUBE O.D. (INCHES)	TUBE MATERIAL	ALLOWABLE WRINKLE HEIGHT (INCHES) ▷	ALLOWABLE OVALITY (% OF NOMINAL O.D) ▷	ALLOWABLE SURFACE MARKS PITS, SCRATCHES (% OF WALL THICKNESS) ▷
LIQUID - 1000 PSI to 3000 PSI -- (INCLUDING RETURN LINES)	ALL	STAINLESS STEEL	0.010	5	5
		AL ALLOY	0.010		
PNEUMATIC - OPERATING PRESSURE OVER 1500 PSI OR TEMPERATURE OVER 160°F	ALL	STAINLESS STEEL	0.010		
ALL OTHER SYSTEMS INCLUDING LIQUID: 0 to 1000 PSI AND PNEUMATIC 0 TO 1500 PSI AND UNDER 160°F	LESS THAN 1.0	STAINLESS STEEL	0.040	10	10
		AL ALLOY	0.020		
	1.0 OR OVER, LESS THAN 2.0	STAINLESS STEEL	0.060	10	10
		AL ALLOY	0.030		
	2.0 OR OVER, LESS THAN 3.0	STAINLESS STEEL	0.080	5	10
		AL ALLOY	0.040		
	3.0 OR OVER	STAINLESS STEEL	0.100	5	10
		AL ALLOY	0.050		

NOTE: Bend radii for tubing shall be in accordance with Boeing Parts Standard BACD2042 or MS33611 unless specifically designated otherwise on drawing.

▷ Replace part when damage exceeds allowable limits.

ASSEMBLY

1. General

A. Engine Configuration

- (1) It is assumed that the engine to be built up is either new or has been overhauled recently. Therefore, the engine configuration should be essentially as follows:
 - (a) All Pratt and Whitney components are installed and properly secured.
 - (b) All engine openings are protected against moisture, dust, and other undesirable elements.
 - (c) The engine is suspended from an overhead monorail or is installed in a suitable buildup stand.
- (2) Engines No. 1 and 2 are basically identical. They differ in nose cowling, disconnect locations for tubing which enters the strut and the striker assemblies on the thrust reversers. Since these differences exist, procedures are given for building the engine up to a maximum neutral configuration (common to both positions on aircraft), then to the final configuration for each position.

B. Equipment

- (1) All special tools and equipment required for power plant buildup are listed in "Special Tools, Fixtures, and Equipment."

C. Procedure

- (1) Each illustration in the section includes a table which keys each step of the assembly to the nomenclature, part number, and required quantity of each part through the use of item numbers. For substitute parts, refer to the Illustrated Parts List Index. The ILLUSTRATED PARTS LIST (Section 1100) contains a complete list of parts, including permitted optional components for all airlines.
- (2) This section is arranged so that engine buildup personnel can initial each assembly step as they proceed through the buildup. Upon completion of the buildup, the text pages can be removed from the manual and kept as a record of the buildup. Locally reproduced text pages may then be inserted in the manual for buildup of next engine.

2. Assembly Practices

A. General

- (1) Make certain that all parts are correctly identified before installation. If a stamped part number cannot be easily read, refer to the Illustrated Parts List or compare the part with one from another engine.
- (2) Unless specified otherwise, tighten all bolts, nuts, and plumbing fittings to standard torque values (Fig. 601 thru 607).
- (3) Unless specified otherwise, lubricate all gaskets, packings (O-rings), and splined shafts prior to installation. See List of Consumable Materials (Fig. 572) for lubricants to be used.
- (4) Take particular care when installing plumbing items to make certain that accumulations of foreign materials are removed from threaded fittings.
- (5) Maintain alignment of tubes with mating fittings to ensure proper installation. Do not tighten tees, elbows, or universal fittings until tubing has been installed.
- (6) When installing hoses, do not allow hose to twist or kink. Make certain that no interference exists between hose and adjacent structure.
- (7) Tighten support clamps only after tubing or hose coupling nuts are tightened.
- (8) Install protective covers over all openings.
- (9) When performing standard electrical bonding test, the specified resistance value should be the overall resistance measured across each bond.

B. Installation of Standard Bolts and Nuts

- (1) The maximum permissible variation of grip length is one size (longer or shorter) as required. A maximum of two standard spacer washers may be used under a nut where necessary to adjust for bolt grip length. To adjust for protruding head where a plate nut is used, a maximum of two standard washers may be used under the bolthead.
- (2) To protect the surface from injury while tightening, one standard spacer washer may be used under either the nut or the bolthead, whichever is being turned.

NOTE 1: A total of three washers is allowed, two for grip and one for surface protection.

NOTE 2: When washers are required, but are not specified in overhaul procedure, material must be similar to that against which washer will bear, i.e., aluminum (AN960D) against aluminum; cadmium plated steel (AN960) against cadmium plated steel; or 5052 aluminum (BACW10T) against magnesium.

- (3) Nuts must not engage incomplete thread next to bolt grip.
- (4) All threads of nut must be engaged and chamfer on threaded end of bolt must be completely clear of nut. Threaded end of bolt or screw must extend at least 1/32 inch through nut.
- (5) Do not use lubricant other than that which is on nut or bolt as supplied by vendor for any installation unless specified in overhaul instructions. (See figure 571.)

Exception: Apply Esnalube 382 to threads of nickel alloy bolts subjected to elevated temperatures up to 2000 degrees F.

- (6) Remove all foreign material from threads of bolt or nut before installation.
- (7) Do not clean nuts unless directed to do so in overhaul instructions.

- (8) Holes and countersinks do not require a protective coating prior to installation of bolts or screws except when they are installed in dissimilar metals, or a protective coating is called out in overhaul instructions or on an engineering drawing.
- (9) Install bolts and screws with heads up or forward, wherever practical, except as noted in overhaul instructions.
- (10) When an NAS1108 (or larger), or NAS1308 (or larger) bolt is used, install a countersunk washer, Specification MS20002 under bolthead.

NOTE: To ensure correct installation of bolt, check grip length. Check for correct orientation of washer. Countersunk side of washer must be in contact with bearing surface of bolthead to allow clearance for radius under bolthead.

- (11) After installation, nuts, and bolt and screw heads must be free of all burrs and sharp edges.

C. Tightening Sequence for Multiple Bolt Installations

- (1) Finger-tighten all bolts or nuts first.
- (2) Snug up opposite bolts or nuts all around.
- (3) Tighten opposite bolts or nuts all around to required torque.

NOTE: Do not tighten adjacent bolts or nuts in succession.

D. Torque Requirements (See figure 601.)

- (1) Use "TORQUE RANGE" columns in figure 601 for all installations not covered by specific drawing and overhaul instructions relating to torque control.
- (2) Torque ranges in figure 601 are for:
 - (a) Clean bolts and nuts as lubricated by the manufacturer.
 - (b) Bolts and nuts surface finished with cadmium plating per Specification QQ-P-416A, type I or type II.

E. Torque Requirements of Self-Locking Nuts (See figure 602.)

- (1) Locking torque is the torque required to start the nut turning, when:
 - (a) The nut is engaged with at least two full bolt threads extending beyond the locking device of the nut.
 - (b) There is no axial load on the nut.
- (2) The minimum to maximum locking torque range, shown in figure 602 is used to determine the usability of a self-locking nut and bolt combination.
- (3) The self-locking mechanism of nuts and nutplates must develop the minimum locking torque at room temperature as shown in figure 602.
- (4) Where there is an apparent excessive torque looseness or torque drag when installing the nut on the bolt, remove the nut and bolt and test for minimum or maximum locking torque according to values in figure 602.

F. Tubing Installation

- (1) Installation of tubing with flareless fittings.
 - (a) Apply thread compound, (figure 571), evenly to entire circumference of male threads and shoulder of flareless sleeve immediately prior to installation except when dri-lubed B-nuts are used. No compound is allowed in interior of tube.
 - (b) Align tube and seat sleeve in fitting before threads are engaged to permit nuts to be threaded easily. Tighten support clamps only after coupling nuts (B-nuts) are tightened.
 - (c) Tighten B-nuts to the torque values specified in figure 603.

- (2) Installation of flared tubing.
 - (a) Align tube with AN818 or equivalent BAC coupling nuts (B-nuts) so that flare will nest on fitting cone with no external restraint other than support clamps. Engage fully by hand. Do not use nut to pull tube into alignment.
 - (b) Align tubes with NAS or BAC fittings which require use of rings (NAS592, NAS595, BACR12H) so that flares will nest on rings without application of external force. Do not use nut to pull tube into alignment. Do not use tool to force engagement of nut.
 - (c) Apply thread compound (selected from figure 571) evenly over entire circumference of male straight threads, back side of flares, and shoulder of sleeves where B-nut is in contact. No compound is allowed on fitting ends, cone surface, inside flare surface, in bore of fittings, interior of tubes, or inside any system components except that petrolatum shall be applied sparingly on the NAS592, NAS595, and BACR12H rings prior to installation. Do not allow petrolatum to enter the tube.
 - (d) Unless specific torques are called out, tighten flared fittings within torque ranges given in figure 604.

G. Hose Installation

- (1) When routing of a hose is not specified, install hose so there will be no kinks or interference between hose and adjacent structure. Bend radii must not be less than the minimum allowed by Specification MS33790 unless otherwise specified.
- (2) When thread compound is used, apply evenly to entire circumference of the male threads only.
- (3) Tighten flared type hose fitting coupling nuts to the torque specified in figure 605. Do not allow hose to twist when installing.
- (4) Tighten flareless type hose coupling nuts in accordance with torques specified in figure 604.
- (5) Flareless type hose coupling nuts which are inaccessible to a torque wrench may be tightened until a distinct increase in torque occurs. Tighten an additional 1/6 to 1/3 turn maximum (one to two hex flats).

H. Installation of Universal Elbow Fittings

- (1) Assemble AN fittings per AND10080. Lubricate with petrolatum, or with Hydraulic Fluid BMS 3-11 or Skydrol Assembly Lube MCS 352 in hydraulic systems which use hydraulic fluid BMS 3-11.
- (2) Assemble NAS fittings per NAS550 and tighten in accordance with torque values of figure 575. Lubricate gasket and bolt threads with petrolatum, or with Hydraulic Fluid BMS 3-11 or Skydrol Assembly Lube MCS 352 in hydraulic systems which use hydraulic fluid BMS 3-11.

I. V-Band Clamp Installation

- (1) Align tube assembly so that mating faces touch making a stable surface and mating tube centerlines are in axial alignment within 0.060 inches.

CAUTION: COUPLINGS ARE NOT SELF-ALIGNING. USE PROPER ALIGNMENT PROCEDURES AND CORRECT TORQUES ON CLAMP NUTS TO ENSURE PERFORMANCE OF THE FINISHED JOINT.

- (2) Place the clamps on the flange.
- (3) Orient the locknut for easy tightening.
- (4) Tighten the clamp locknut to 5 lb-in. above the running torque. Make sure that alignment is proper by tapping the entire circumference of clamp lightly with a wood, leather or soft plastic mallet.
- (5) Retighten the clamp locknut to the correct torque. (Ref. Figure 606.)

J. Electrical Wiring Installation

NOTE: Refer to Fig. 606, 607 for torque values of clamp nuts and electrical ground connections.

- (1) Tie wires leading to a common connector together to form a single cable where practical.
- (2) Several cable bundles may be routed in the same raceway or held by one clamp. Several cables can be tied together only when necessary for bundle integrity or mechanical strength.
- (3) Maintain a 6-inch space between wires and lines carrying flammable fluids wherever possible. Minimum acceptable spacing is 2 inches.
- (4) Maintain minimum spacing between unprotected wire and heated equipment, such as thermal anti-icing or air conditioning ducts, as follows:
 - (a) Standard Wire -- minimum clearance of one inch from side or bottom and two inches from top.
 - (b) High Temperature Wire -- minimum clearance of one-half inch from side or bottom and one inch from top.
- (5) Support wires so that there is a minimum clearance of 0.25 inch between wire and any sharp edge.
- (6) Clip-on nuts (BACN10F, for use with 10-32 screws) are optional to standard hexagonal self-locking nuts when used in accordance with Specification BACN10FX.
- (7) Install all connectors so that weight of connector is not supported by wire, or wires terminating in the connector.
- (8) Grounding ac and dc current to a common terminal is not permitted.
- (9) Do not connect current return conductors to parts made of magnesium.
- (10) A maximum resistance of 0.0006 ohms is allowed for generator ground (generator neutral).
- (11) Lockwire all connectors (except bayonet type) which are in areas not normally accessible for periodic maintenance.
- (12) Remove unused ground screws from connectors.

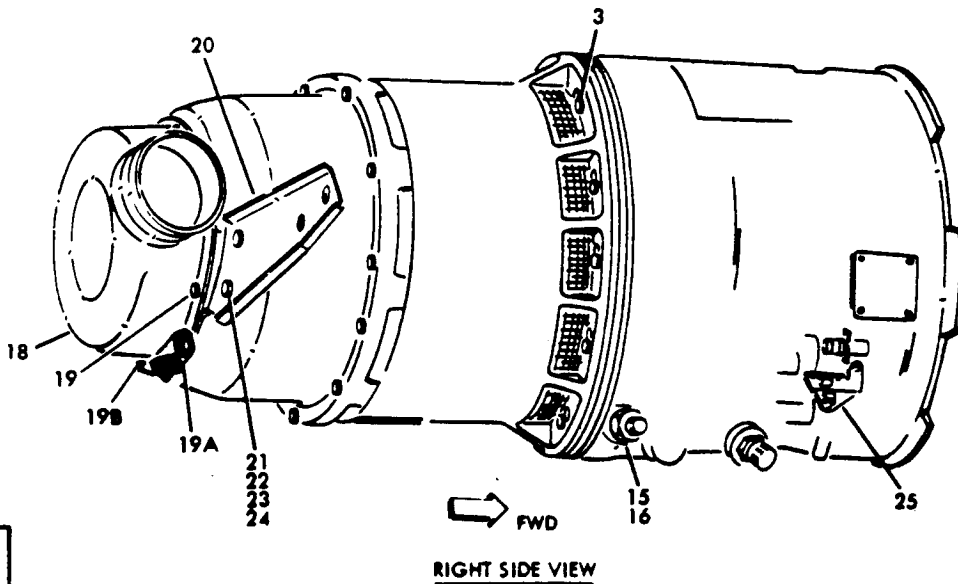
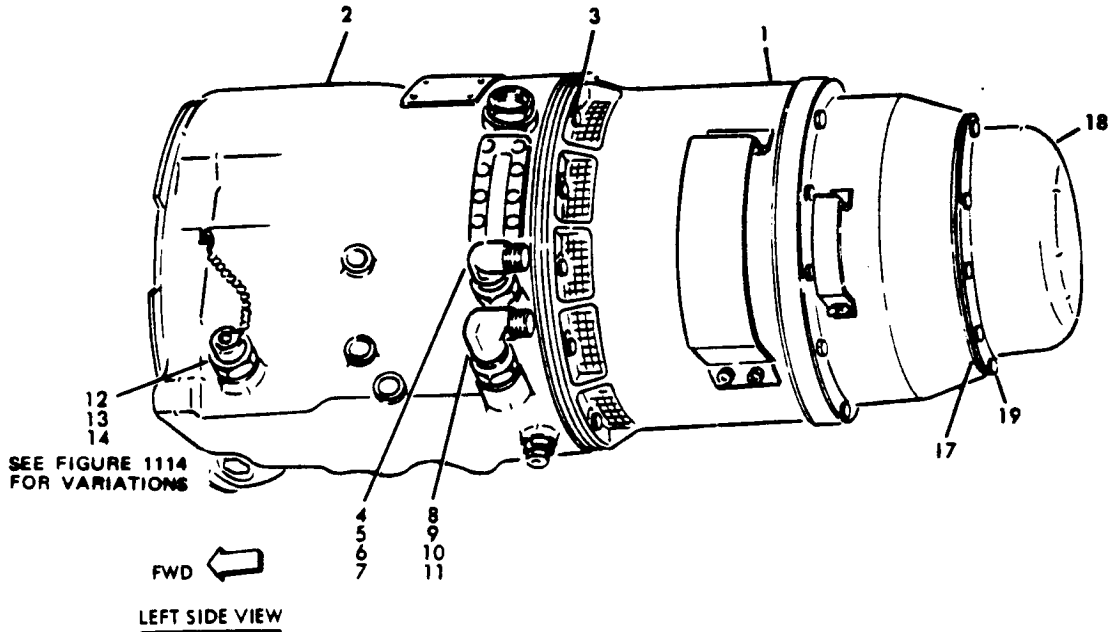
3. Preassembly of Components

NOTE: Buildup of an engine may be facilitated by preassembling certain QEC kit components which go together to form an easily installed unit. Components recommended for preassembly are as follows:

PREASSEMBLED COMPONENTS	PARAGRAPH	FIGURE
CSD and AC Generator Buildup 65-50571	3.A.	501
CSD and AC Generator Buildup 65-21931	3.B.	501A
AC Generator Cooling System Components	3.C.	502
CSD Oil Filter and System Components	3.D.	503
CSD Oil Cooler and System Components	3.E.	504
Engine Oil Pressure Transmitter and Switch	3.F.	505
8th Stage Air Supply Manifold	3.G.	506
13th Stage High Pressure Manifold	3.G.	507
Cabin Air Manifold	3.H.	508
Engine Starter Manifold	3.H.	509
Air-Conditioning Precooler Buildup	3.I.	510
Engine Starter Buildup	Deleted	
Engine Oil Differential Pressure Switch	3.K.	512
Differential Pressure Regulator	3.L.	513
Hydraulic Pump Buildup (Kellogg)	3.M.	514
Hydraulic Pump Buildup (Vickers)	3.M.	515
Hydraulic Filter Buildup	3.N.	516
Nose Cowl TAI Filter and Valve	3.O.	517
Nose Cowl TAI Thermostatic Valve	3.O.	518
Engine Drain Fitting Assemblies	3.P.	519

- A. 65-50571 -- Constant speed drive (CSD), AC generator, and generator cooling duct (Fig. 501)
- (1) Lubricate input shaft spline of AC generator (1) with a mixture of molybdenum disulfide MIL-M-7866, and Texaco Unitemp 500 grease. Mix lubricants in equal parts by weight. Premixed lubricant, Sundstrand 688272-2, may be substituted for mixture.
 - (2) Attach AC generator to CSD (2) with twelve nuts (3) as follows:
 - (a) Install nuts (3) on CSD output flange studs, leaving a gap of approximately 0.50 inch between flange and nut.
 - (b) Engage external generator drive spline with CSD internal drive spline while allowing nuts (3) to enter large holes in mounting flange of generator.
 - (c) Move generator mounting flange up against CSD output flange; then rotate generator clockwise (looking from rear) until stud shanks reach end of slots in generator mounting flange.
 - (d) Tighten nuts (3) 160-190 lb-in.
 - (3) Install jamnut (4) on elbow (5), and install elbow in OIL IN boss on lower left side of CSD with O-ring (6) and backup ring (7). Do not tighten jamnut (4).
 - (4) Install jamnut (8) on elbow (9), and install elbow in OIL OUT boss on lower left side of CSD with O-ring (10) and backup ring (11). Do not tighten jamnut (8).
 - (5) Install coupling half (12) and O-ring (13) on CSD PAD CAVITY FILL boss.
 - (6) Install dust cover (14) on coupling half (12), and attach chain of cover to lug on CSD case. Crimp ring on chain shut after installation.

- (7) Install union (15) and O-ring (16) in OUTPUT SEAL DRAIN boss on lower right side of CSD.
- (8) Attach duct support plate (17), air blast connector (18) and bracket assembly (19A) to aft end of AC generator using screws (19 and 19B).
- (9) Lockwire all screws (19 and 19B).
- (10) Insert bolts (21) through attachment holes of bracket (20), place washer (22) on each bolt, and fasten bracket (20) to duct support plate (17) with washers (23) and nuts (24).
- (11) Remove existing screws and clamp. Install bracket (25), and reinstall clamp and screws.
- (12) Cover or cap all openings in assembly for protection against dust and dirt.



REF.	FIG
522	
523	
1114	
1122	

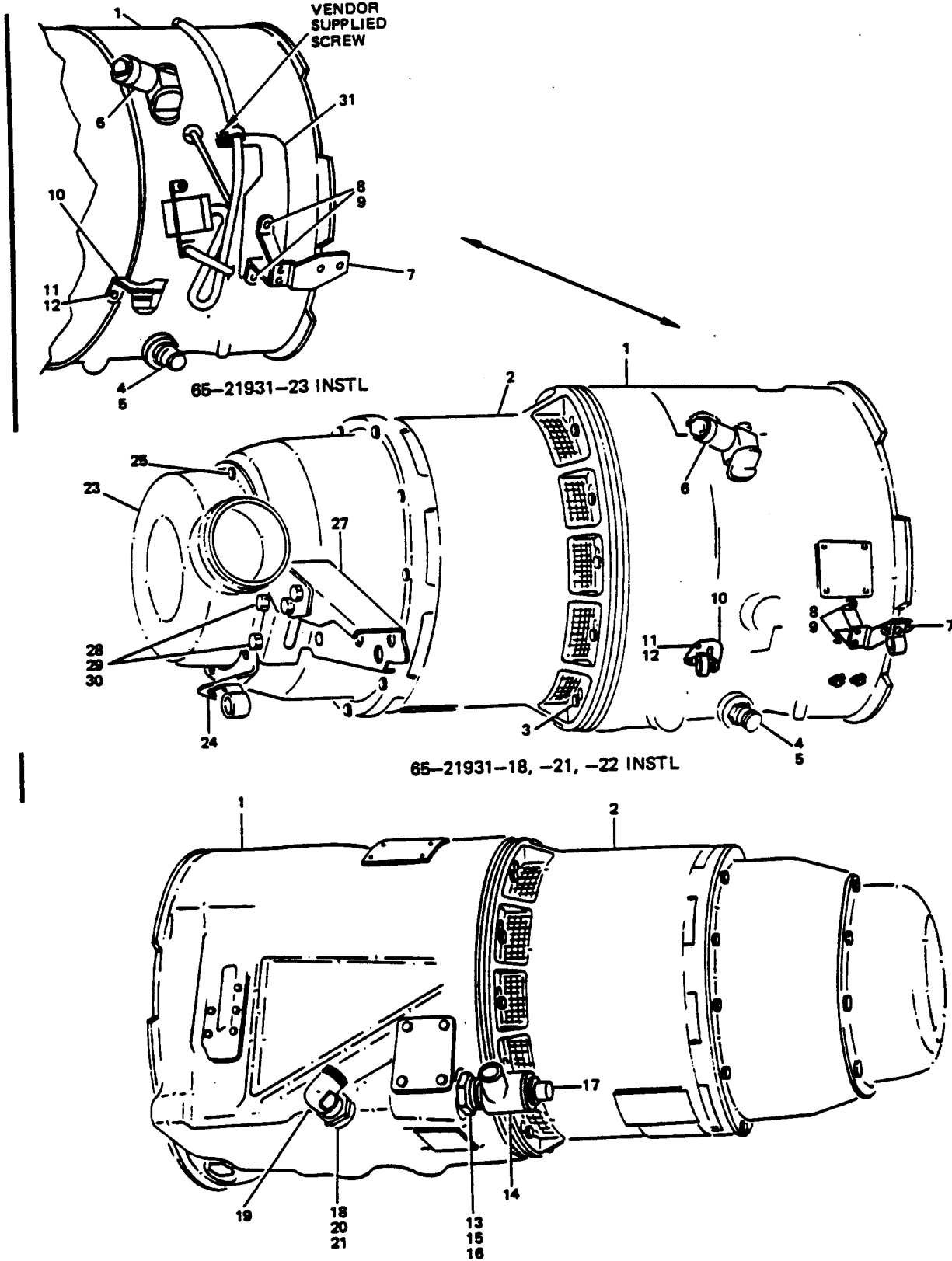
CSD and AC Generator Buildup
 Figure 501 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
B.(1)	1	AC GENERATOR	10-61224-10		1
(2)	2	CONSTANT-SPEED DRIVE	10-61223-2		1
	3	NUT	BACN10BL6L		12
	3	NUT	BACN10GW6		12
(3)	4	JAMNUT	AN6289-10		1
	5	ELBOW, 90° BULKHEAD	MS21908-10		1
	6	O-RING	BACP11P-B10		1
	7	RING, BACKUP	MS28777-10		1
(4)	8	JAMNUT	AN6289-8		1
	9	ELBOW, 90° BULKHEAD	MS21908-8		1
	10	O-RING	BACP11P-B8		1
	11	MS28777-8	RING, Backup		1
(5)	12	COUPLING HALF	OMP2506-2		1
	13	O-RING	BACP11P112		1
(6)	14	COVER, DUST	OMP2506-4		1
(7)	15	UNION	MS21902-6		1
	16	O-RING	BACP11P112		1
(8)	17	PLATE, DUCT SUPPORT	69-43285-1		1
	18	CONNECTOR, AIR BLAST	938D844-3		1
	19	SCREW	AN501AD10-7		3
	19A	BRACKET ASSEMBLY	69-52217-2		1
	19B	SCREW	AN501AD10-8		2
(9)		NO PARTS REQUIRED			
(10)	20	BRACKET	69-43284-6		1
	21	BOLT	NAS1103-3		2
	22	WASHER	AN960C10		2
	23	WASHER	AN960C10L		2
	24	NUT	NAS679A3W		2
(11)	25	BRACKET ASSEMBLY	69-52217-1		
(12)		NO PARTS REQUIRED			

B. 65-21931 -- Constant speed drive, AC generator, and generator cooling duct as follows: (Fig. 501A)

- (1) Lubricate spline of generator input shaft with 1:1 mixture (by weight) of molybdenum disulfide (MIL-M-7866), and Texaco Unitemp 500 grease. Premixed Lubricant, Sundstrand 688272-2 may be used as a substitute.
- (2) Attach constant speed drive (1) to AC generator (2) with nuts (3). Tighten nuts (3) to 160-190 lb-in.
- (3) Install union (5) with O-ring (4) in input pad drain port and tighten.
- (4) Install vented cap (6) and lockwire per 20-50-02 double twist method.
- (5) On 65-21931-23 installation, remove existing vendor supplied screw from wire bundle clamp. Install cover plate (31) under wire bundle clamp and reinstall screw thru clamp and cover plate.
- (6) Install bracket (7) using washer (8) and screw (9).
- (7) Install bracket (10) using washer (12) and screw (11).
- (8) Install nut (13) or bulkhead fitting (14).
- (9) Install bulkhead fitting (14) with washer (15) and O-ring (16) in OIL-IN port. Do not tighten nut (13). Install switch (17) on bulkhead fitting (14).
- (10) Install nut (18) on elbow (19).
- (11) Install elbow (19) in OIL-OUT port with washer (20) and O-ring (21). Do not tighten nut (18).
- (12) Attach duct support plate (22), air blast connector (23) and bracket (24) to aft end of AC generator (2) using screws (25, 26). Lockwire all screws per 20-50-02.
- (13) Install bracket (27) to support plate (22) using bolts (28) washer (29) and nut (30).

OVERHAUL MANUAL



CSD and AC Generator Buildup
Figure 501A (Sheet 1)

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(1,2)	1	CONSTANT SPEED DRIVE ASSY	731753		1
	1	CONSTANT SPEED DRIVE ASSY	735511A		1
	1	CONSTANT SPEED DRIVE ASSY	735511		1
	2	AC GENERATOR ASSY	976J598-1		1
	2	AC GENERATOR ASSY	976J449-1		1
	2	AC GENERATOR ASSY	976J498-2		1
	3	NUT	BACN10BL6L		12
(3)	4	O-RING	BACP11PB6		1
	5	UNION	BACU24K6		1
(4)	6	VENTED CAP	706183		1
(5)	31	COVER PLATE	69-74656-1		1
(6)	7	BRACKET ASSY	69-71727-2		1
	8	WASHER	AN960C10L		2
	9	SCREW	NAS1352C3-10		2
(7)	10	BRACKET ASSY	69-71727-1		1
	10	BRACKET ASSY	69-71727-9		1
	10	BRACKET ASSY	69-71727-10		1
	11	SCREW	NAS1352C3-8		2
	12	WASHER	AN960C10L		2
(8)	13	NUT	AN6289-10		1
	14	BULKHEAD FITTING	69-71573-1		1
(9)	15	WASHER	MS28773-10		1
	16	O-RING	M83248-2-910		1
	17	SWITCH, THERMAL	55584		1
(10)	18	NUT	AN6289-8		1
	19	ELBOW	MS21908-8		1
(11)	20	WASHER	MS28773-8		1
	21	O-RING	M83248-2-908		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(12)	22	DUCT SUPPORT PLATE	69-43285-1		1
	23	AIR BLAST CONNECTOR	65C20874-1		1
	24	BRACKET ASSY	69-52217-2		1
	25	SCREW	NAS1802-3D7		3
	26	SCREW	NAS1802-3D8		2
(13)	27	BRACKET ASSY	69-43284-7		1
	28	BOLT	NAS1103-3		2
	29	WASHER	AN960C10L		2
	30	NUT	BACN10JC3		2

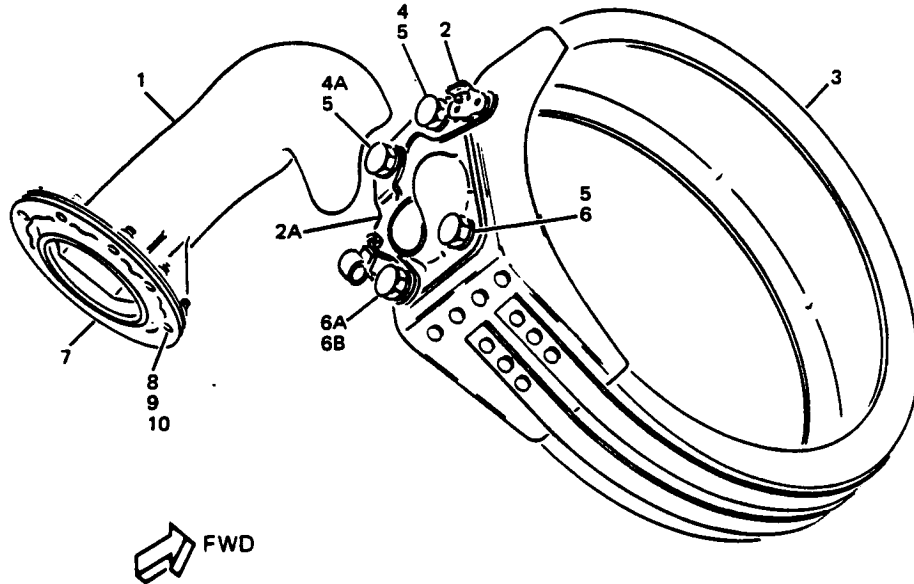
CSD and AC Generator Buildup
Figure 501A (Sheet 3)

C. Preassemble components of the AC generator cooling air system as follows (Fig. 502):

- (1) Attach exhaust duct fitting (1) and bracket assemblies (2 and 2A) to exhaust collector ring (3) with bolts (4, 4A, 6 and 6A) and washers (5).
- (2) Install seal assembly (7) with bolts (8), washers (9), and nuts (10).
- (3) Install covers over all duct openings in assembly.

737
POWERPLANT BUILDUP

OVERHAUL MANUAL



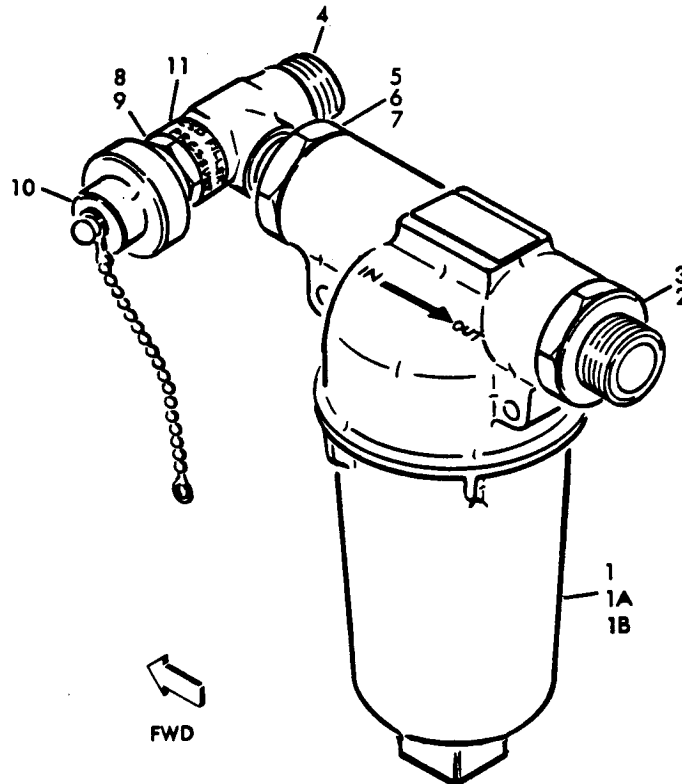
REF.
 FIG
 523
 1122

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
C. (1)	1	FITTING-EXHAUST	65-46505-3		1
	1	FITTING-EXHAUST	65-46505-1		1
	2	BRACKET ASSEMBLY	69-45402-1		1
	2A	BRACKET ASSEMBLY	69-71727-8		1
	2A	BRACKET ASSEMBLY	69-52217-3		1
	3	COLLECTOR RING	65-22429-22		1
	3	COLLECTOR RING	65-22429-19		1
	3	COLLECTOR RING	65-22429-15		1
	4	BOLT	NAS1103-7		1
	4	BOLT	NAS1103-6		1
	4A	BOLT	NAS1103-8		1
	4A	BOLT	NAS1103-7		1
	5	WASHER	AN960C10L		3
	6	BOLT	NAS1103-7		1
	6	BOLT	NAS1103-6		1
	6	BOLT	NAS1103-5		1
	6A	BOLT	BACB30LU3-6		1
	6A	BOLT	NAS1103-7		1
	6B	WASHER	AN960C10L		1
	(2)	7	SEAL ASSEMBLY	69-31696-1	
8		BOLT	BACB30LU3-2		8
8		BOLT	BACB30FL3-2		8
9		WASHER	AN960-10L		8
10		NUT	BACN10JC3		8
10		NUT	NAS679A3W		8
(3)		NO PARTS REQUIRED			

AC Generator Cooling System Components
 Figure 502

D. Assemble components of CSD oil pressure system with CSD oil filter as follows (Fig. 503):

- (1) Install new filter element (1A) and new packing (1B) in filter (1). Install union (2) and O-ring (3) in OUT port of filter (1). Tighten union.
- (2) Install jamnut (5) on tee (4), and install tee (4) in IN port of filter (1) with O-ring (6) and backup ring (7). Do not tighten jamnut (5).
- (3) Install quick-disconnect coupling (8) on long end of tee (4) with O-ring (9).
- (4) Install dust cover (10) on quick-disconnect coupling (8).
- (5) Install Metal-Cal (11) on upper side of tee (4) adjacent to coupling (8) per 20-50-05.
- (6) Install caps or plugs in all open ports.

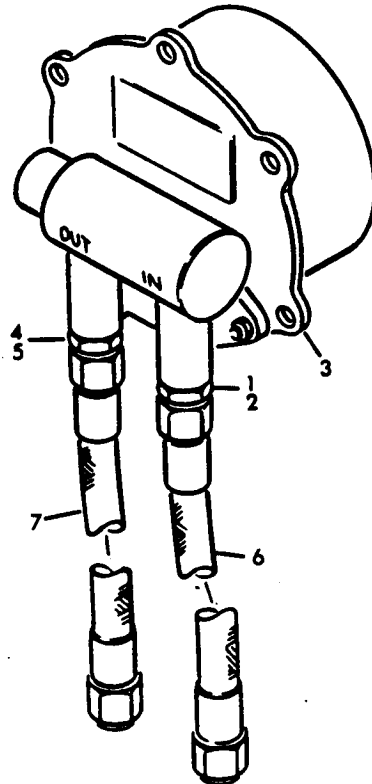


REF.
FIG
549
1123

CSD Oil Filter and System Components
Figure 503 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
D.(1)	1	FILTER	7581418		1
	1	FILTER	7586890		1
	1A	FILTER ELEMENT	7513182		1
	1A	FILTER ELEMENT	7583141		1
	1B	PACKING	MS29561-228		1
	1B	PACKING	M83248-1-228		1
	2	UNION	MS21916-12-8		1
	3	O-RING	BACP11PB12		1
	(2)	4	TEE	118-10560	
5		JAMNUT	AN6289-12		1
6		O-RING	BACP11PB12		1
7		RING, BACKUP	MS28777-12		1
(3)	8	COUPLING, QUICK-DISCONNECT	OMP2506-2		1
	9	O-RING	BACP11PB6		1
(4)	10	COVER, DUST	OMP2506-4		1
(5)	11	METAL-CAL	BACM10P6GU		1
(6)		NO PARTS REQUIRED			

- E. Assemble components of the CSD oil cooling system with CSD oil cooler as follows: (See figure 504.)
- (1) Install union (1) and O-ring (2) in IN port of CSD oil cooler assembly (3). Tighten union.
 - (2) Install union (4) and O-ring (5) in OUT port of CSD oil cooler assembly (3).
 - (3) Attach hose (6) to union (1). Tighten hose end nut.
 - (4) Attach hose (7) to union (4). Tighten hose end nut.
 - (5) Plug or cap all open ports in assembly.



REF.
FIG
550
1120
1135

FWD 

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
E.(1)	1	UNION	MS21902-8		1
	2	O-RING	BACP11P-B8		1
	3	COOLER ASSEMBLY, OIL	10-61233-1		1
	3	COOLER ASSEMBLY, OIL	10-61233-9		1
(2)	4	UNION	MS21902-10		1
	5	O-RING	BACP11P-B10		1
(3)	6	HOSE	BACH30AS08F0173 *[1]		1
	6	HOSE	BACH5R073DDF		
(4)	7	HOSE	BACH30AS10F0173 *[2]		1
	7	HOSE	BACH5R073EEF		
(5)		No parts required			

*[1] BACH5R0173DDF REPLACES BACH30AS08F0173
*[2] BACH5R0173EEF REPLACES BACH30AS10F0173

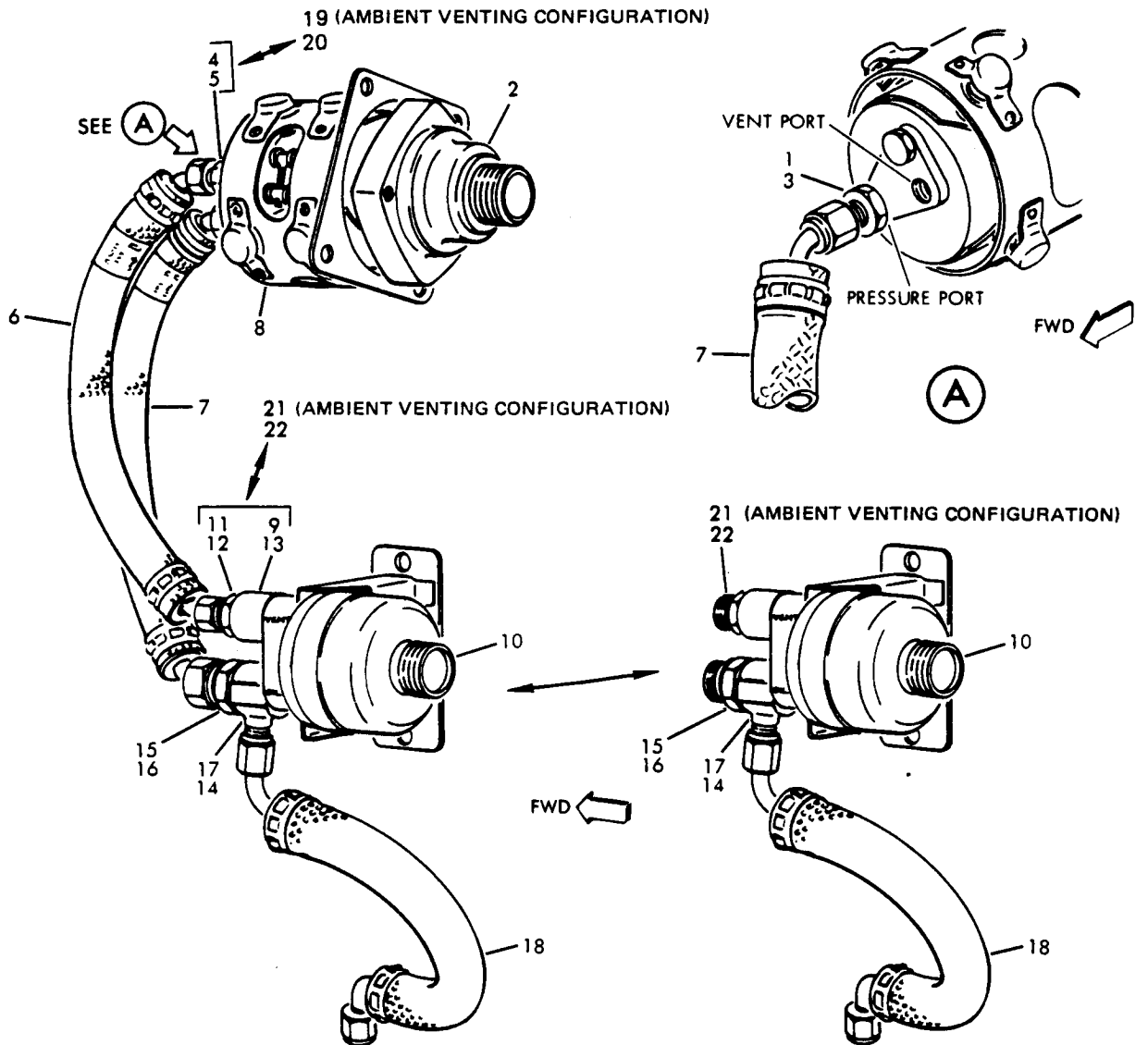
OVERHAUL MANUAL

F. Assemble components of engine oil pressure system with oil pressure transmitter and pressure switch as follows (Fig. 505):

- (1) On 65-50882-1, -2 installations, build up oil pressure transmitter as follows:
 - (a) Install reducer union (1) in PRESSURE port of oil pressure transmitter (2) with O-ring (3). Tighten union.
 - (b) On installations having transmitter vented to engine gearbox:
 - 1) Install union (4) with O-ring (5) in VENT hole of transmitter (2).
 - 2) Attach hose (6) to union (4). Do not tighten.
 - (c) On installations having transmitter vented to ambient pressure:
 - 1) Install vent port (19) with O-ring (20) to VENT hole of transmitter.
 - (d) Attach end fitting of hose (7) to union (1). Do not tighten end fitting.
 - (e) Install oil pressure transmitter (2) in shock mount (8). Tighten T-bolt in shock mount finger-tight only. Position of transmitter must be adjusted when installed on engine.
- (2) On 65-50882-1, -2, -3 installations, build up oil pressure switch as follows:
 - (a) On installations having pressure switch vented to engine gearbox:
 - 1) Install elbow (9) with bolt (11) and O-rings (12, 13) in VENT hole of switch (10). Do not tighten bolt.
 - (b) On installations having pressure switch vented to ambient pressure:
 - 1) Install vent port (21) with O-ring (22) in VENT hole of switch.
 - (c) Install elbow (14) in OIL port of pressure switch with universal bolt (15) and O-rings (16, 17). Tighten universal bolt finger-tight only.
 - (d) Attach end of hose (18) to side port of elbow (14). Do not tighten end fitting of hose.

OVERHAUL MANUAL

- (3) On 65-50882-1, -2 installations:
 - (a) Attach end fitting of hose (7) to universal bolt (15). Do not tighten end fitting.
- (4) On 65-50882-1, -2 installations having pressure switch vented to engine gearbox:
 - (a) Attach hose (6) to bolt (11). Do not tighten.
- (5) Cap or plug all openings and electrical connections in assembly.



REF.
FIG
543
1115
1126

65-50882-1, -2
INSTALLATION

65-50882-3 INSTALLATION
(TRANSMITTER NOT SHOWN, REFER TO FIG. 543)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
F.					
(1)(a)	1	UNION, REDUCER	MS21916-6-4		1
	2	TRANSMITTER, OIL PRESSURE	ST104AB		1
	2	TRANSMITTER, OIL PRESSURE	ST104Z		1
	3	O-RING	BACP11PB4		1
(b)	4	UNION	MS21902-4		1
	5	O-RING	BACP11PB4		1
	6	HOSE	10-61357-2		1
(c)	19	VENT PORT	66-13955-875		1
	20	O-RING	BACP11PB4A		1
(d)	7	HOSE	10-61357-4		1
(e)	8	SHOCKMOUNT	K710-22		1
(2)(a)	9	ELBOW, UNIVERSAL 90°	NAS1237-4		1
	10	SWITCH, PRESSURE	42D107A1		1
	11	BOLT, UNIVERSAL FITTING	NAS1236-4H		1
	12	O-RING	BACP11PB4		1
	13	O-RING	BACP11P11		1
(b)	21	VENT PORT	66-13955-875		1
	22	O-RING	BACP11PB4A		1
(c)	14	ELBOW	66-19214-1		1
	15	BOLT, UNIVERSAL FITTING	69-45458-1		1
	16	O-RING	BACP11PB4		1
	16	O-RING	BACP11PB4A		1
	17	O-RING	BACP11P11		1
(d)	18	HOSE	10-61357-3		1
(3)		NO PARTS REQUIRED			
(4)		NO PARTS REQUIRED			
(5)		NO PARTS REQUIRED			

Engine Oil Pressure Transmitter and Switch
Figure 505 (Sheet 2)

G. Preassemble components of the air-conditioning system into two subassemblies as follows:

(1) Assemble components of 8th-stage air supply manifold as follows:
(See figure 506.)

(a) Attach manifold assembly (1) to duct assembly (2) with wear ring (3) and shims (4 and 5) as follows:

1) Install wear ring (3) in bearing nut of duct assembly (2), and measure gap between ends of ring. Gap must not exceed 0.030 inch.

NOTE: Ends of wear ring may be filed if necessary to facilitate installation.

2) Join duct assembly (2) and manifold assembly (1) with wear ring (3) in place, but without shims. Tighten bearing nut on duct assembly (2) to 70-90 pound-inches.

3) Measure gap between bearing nut and retainer rim with feeler gage. (See gap X in detail A.)

4) Add 0.012 to 0.014 inch to gap dimension to obtain total combined thickness of shims (4 and 5) to be used for final assembly of manifold and duct. Adjust thickness of shims by removing laminations from shim (4.)

NOTE: Deleted.

5) Detach duct and manifold assemblies.

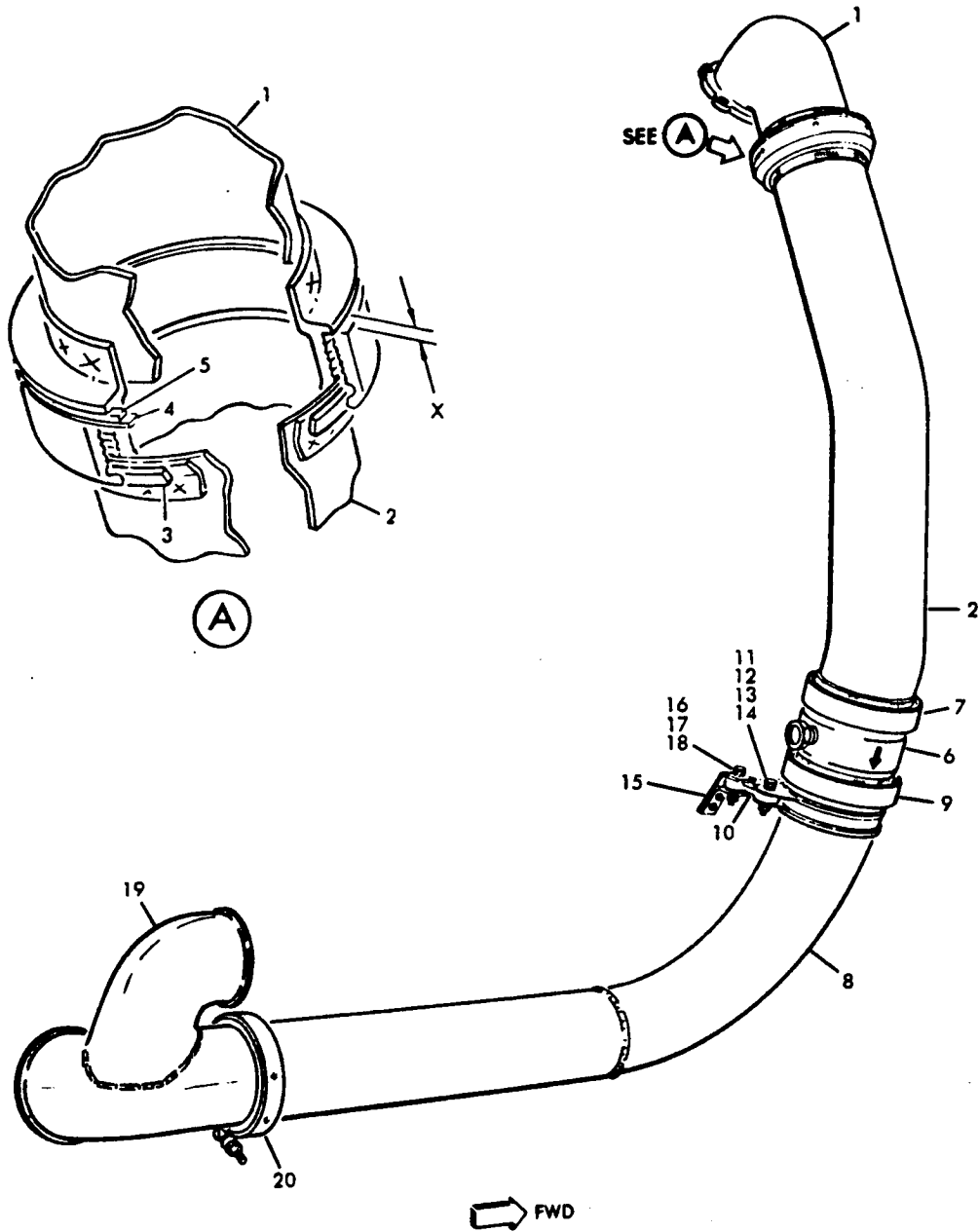
6) Coat threads of bearing nut and retainer with antiseize compound, Specification MIL-A-907. Reassemble duct and manifold assemblies with wear ring (3) and shims (4 and 5) in place.

NOTE: Edges of shims must be flush with outer surface of bearing nut.

7) Tighten bearing nut to 380-400 pound-inches.

8) Check joint for freedom of movement, and lockwire nut to retainer per Subject 20-50-02.

- (b) Attach check valve (6) to free end of duct assembly (2) with clamp (7). Orient valve as shown in figure 506.
- (c) Attach duct assembly (8) to check valve (6) with clamp (9).
- (d) Install link (10) in bracket on duct assembly (8) with bolt (11), washers (12) and washer (13). Place washer (12) on each side of link. Install nut (14).
- (e) Install bracket assembly (15) on free end of link (10) with bolt (16), washer (17), and nut (18).
- (f) Attach duct assembly (19) to free end of duct assembly (8) with clamp (20).



REF.
FIG
528
1111

8th Stage Air Supply Manifold
Figure 506 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
G	(1)(a)	1	MANIFOLD ASSEMBLY	69-45174-1	1
		2	DUCT ASSY, 8TH-STAGE	65-53865-11 (OPT)	1
		2	DUCT ASSEMBLY	65-53865-1 (OPT)	1
		3	RING, WEAR	69-26229-2 (OPT)	1
		3	RING, WEAR	69-64750-3 (OPT)	1
		4	SHIM, Laminated	69-56508-3	1
		5	SHIM, Solid	69-56508-6	1
	(b)	6	VALVE, Check	10-60508-3	1
		7	CLAMP	BACC10DU350Y	1
	(c)	8	DUCT ASSEMBLY	65-53865-802	1
		8	DUCT ASSEMBLY	65-53865-803 (optional)	1
		9	CLAMP	BACC10DU350Y	1
	(d)	10	LINK	66-14598-1	1
		11	BOLT	NAS1104-8	2
		12	WASHER	AN960416L	2
		13	WASHER	AN960416	1
		14	NUT	BACN10JC4	1
	(e)	15	BRACKET ASSEMBLY	69-43437-1	1
		16	BOLT	NAS1104-8	1
		17	WASHER	AN960C14	1
18		NUT	BACN10JC4	1	
(f)	19	DUCT	65-53866-16 (PREF)	1	
	19	DUCT	65-53866-800 (OPT)	1	
	20	CLAMP	BACC10DU300Y	1	

(2) Assemble components of 13th-stage high pressure manifold assembly as follows: (See figure 507.)

(a) Attach manifold assembly (1) to duct assembly (2) with wear ring (3) and shims (4 and 5) as follows:

1) Install wear ring (3) in bearing nut of duct assembly (2), and measure gap between ends of ring. Gap must not exceed 0.030 inch.

NOTE: Ends of wear ring may be filed if necessary to facilitate installation.

2) Join manifold assembly (1) and duct assembly (2) with wear ring (3) in place, but without shims. Tighten bearing nut on duct assembly (2) to 70-90 pound-inches.

3) Measure gap between bearing nut and retainer rim with feeler gage. (See gap X in detail A.)

4) Add 0.012 to 0.014 inch to gap dimension to obtain total combined thickness of shims (4 and 5) to be used for final assembly of manifold and duct. Adjust thickness of shims by removing laminations from shim (4).

NOTE: Deleted.

5) Detach duct and manifold assemblies.

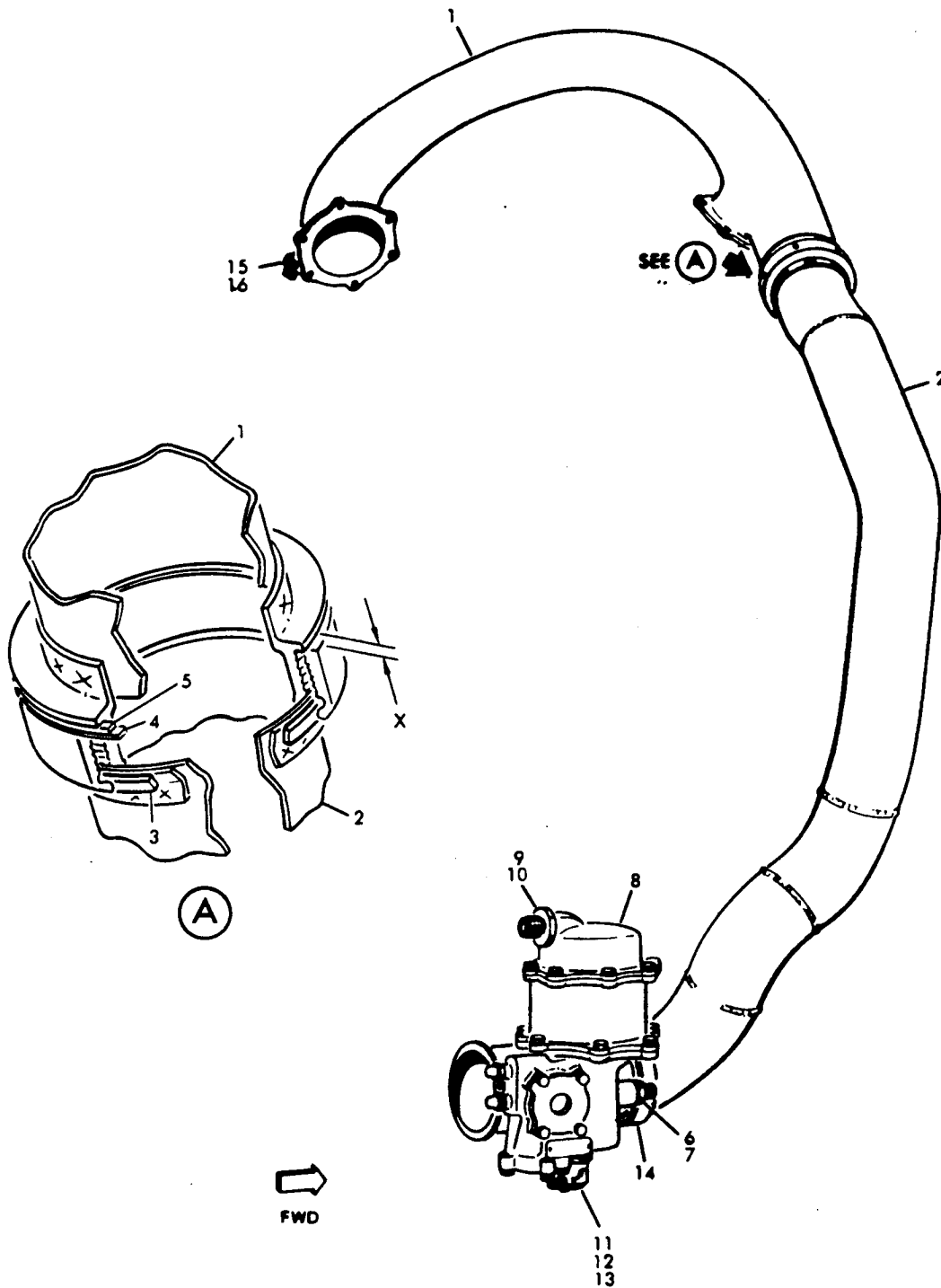
6) Coat threads of bearing nut and retainer with antiseize compound, Specification MIL-A-907. Reassemble duct and manifold assemblies with wear ring (3) and shims (4 and 5) in place.

NOTE: Edges of shims must be flush with outer surface of bearing nut.

7) Tighten bearing nut to 380-400 pound-inches.

8) Check joint for freedom of movement. Lockwire nut to retainer per Subject 20-50-02.

- (b) Install elbow (6) and gasket (7) in shutoff valve (8). Do not tighten elbow.
- (c) Install elbow (9) and gasket (10) in shutoff valve (8). Do not tighten elbow.
- (d) Install jamnut (12) on elbow (11), and install elbow in shutoff valve (8) with gasket (13). Do not tighten jamnut.
- (e) Attach shutoff valve (8) to free end of duct assembly (2) with clamp (14).
- (f) Install union (15) and gasket (16) in boss on left end of manifold assembly (1). Tighten union.



REF.
FIG
52B
1111

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(2)(a)	1	MANIFOLD ASSY	65-22441-39		1
	1	MANIFOLD ASSY	65-22441-34		1
	1	MANIFOLD ASSY	65-22441-31		1
	1	MANIFOLD ASSY	65-22441-21 *[1]		1
	1	MANIFOLD ASSY	65-22441-28 (used on engines with vortex dissipator)		1
	2	DUCT ASSY	65-53864-800		1
	3	RING, WEAR	69-26230-2 (OPT)		1
	3	RING, WEAR	69-64750-2 (OPT)		1
	4	SHIM, LAMINATED	69-56508-2		1
	5	SHIM, SOLID	69-56508-5		1
(b)	6	UNION	MS21902-6		1
	7	GASKET	BACP11P-B6		1
	8	VALVE, SHUTOFF	10-60492-4		1
(c)	9	UNION	MS21902-6		1
	10	GASKET	BACP11P-B8		1
(d)	11	ELBOW	MS21908-6		1
	12	JAMNUT	AN924-6		1
	13	GASKET	BACP11P-B6		1
(e)	14	CLAMP	BACC10DU300Y		1
(f)	15	UNION	ER 10339		1
	16	GASKET	AN901-4C		1

*[1] Optional part

*[2] Replaces 65-22441-21 on engine retrofitted per 65-75985

13th-Stage High Pressure Manifold
Figure 507 (Sheet 2)

H. Assemble components of cabin air and engine starter manifolds in two subassemblies as follows:

(1) Cabin air manifold components (Fig. 508).

- (a) Install thermostat valve (1) and gasket (2) in welded boss on duct assembly (3 or 3A) with bolts (4), washers (5), and nuts (6). For thermostat valve (1) optional installation, rotate valve 180 degrees from noted installation.
- (b) Install union (7) and gasket (8) in port of thermostat valve (1).
- (c) Install retainer (10) on shutoff valve (11) with clamp (12).
- (d) If duct assembly (3A) is used, attach to retainer (10) on shutoff valve (11) as follows:

NOTE: Install valve with electrical connector down.

- 1) Install wear ring (13) in bearing nut of duct assembly (3A), and measure gap between ends of ring. Gap must not exceed 0.030 inch.

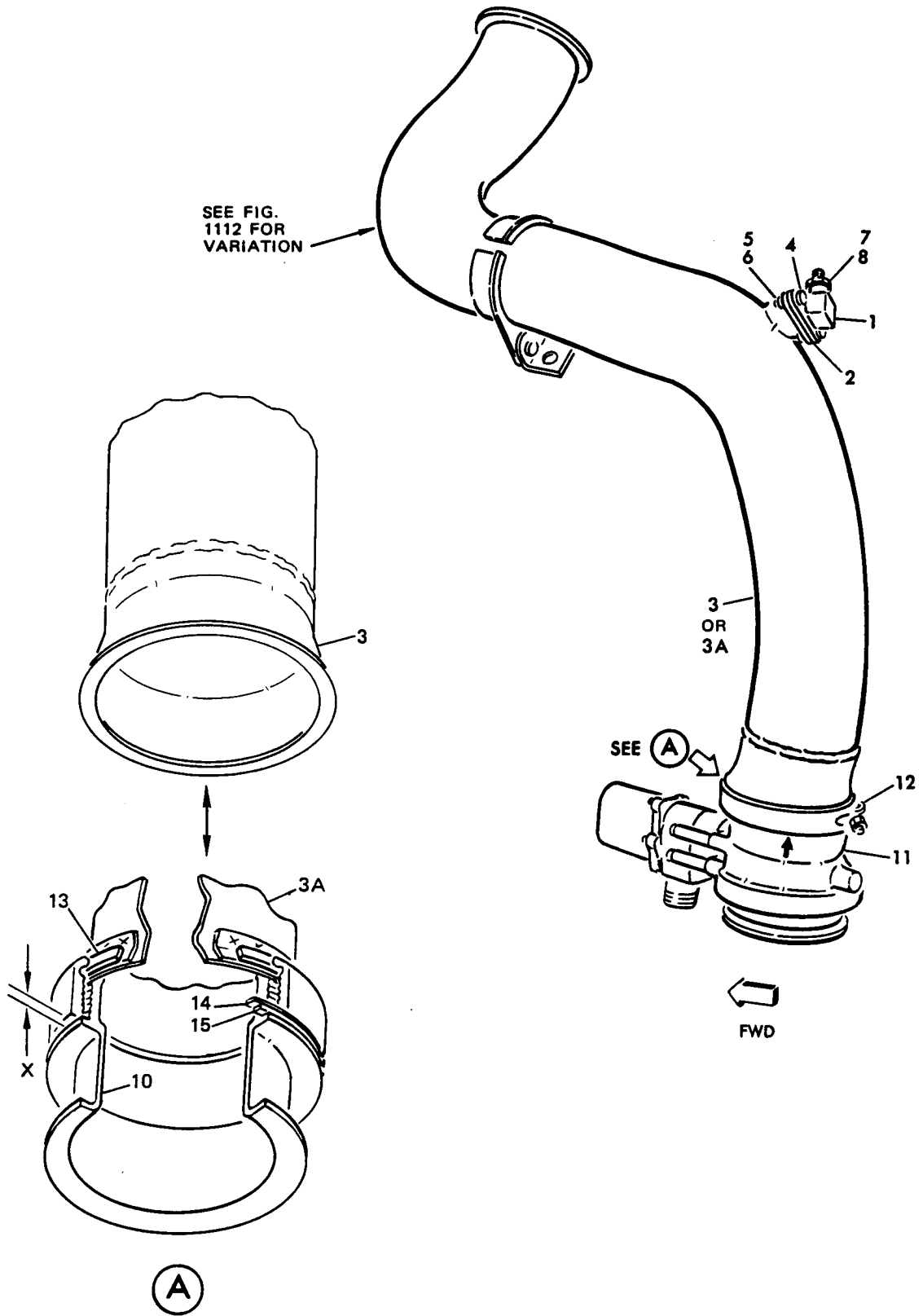
NOTE: Ends of wear ring may be filed if necessary to facilitate installation.

- 2) Join duct assembly (3A) and shutoff valve (11) with wear ring (13) in place, but without shims. Tighten bearing nut on duct assembly (3A) to 70-90 lb-in.
- 3) Measure gap between bearing nut and rim of retainer on valve with a feeler gage. (See gap X in detail A.)
- 4) Add 0.012 - 0.014 inch to gap dimension to obtain total combined thickness of shims (14, 15) to be used for final assembly of duct and valve. Adjust thickness of shims by removing laminations from shim (14).
- 5) Detach valve from duct assembly.

- 6) Coat threads of bearing nut and retainer with anti-seize compound, MIL-A-907. Reassemble duct assembly and valve with wear ring (13) and shims (14 and 15) in place.

NOTE: Edges of shims must be flush with outer surface of bearing nut.

- 7) Tighten bearing nut to 380-400 lb-in.
 - 8) Check joint for freedom of movement. Lockwire nut to retainer per 20-50-02.
- (e) Cap open ends of ducts.



REF.
 FIG
 509
 547
 1112

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
H.					
(1)(a)	1	VALVE, THERMOSTATIC	129240-1-1		1
	2	GASKET	856411-1		1
	3	DUCT ASSY	65-53863-30		1
	3A	DUCT ASSY	65-53863-802		1
	4	BOLT	NAS1103-8		2
	5	WASHER	AN960-10L		2
	6	NUT	BACN10JC3		2
(b)	7	UNION	MS21902-4		1
	8	GASKET	AN901-4C		1
(c)	10	RETAINER	69-17969-3		1
	11	VALVE, SHUTOFF	2760000-101		1
	11	VALVE, SHUTOFF	10-60495-6		1
	12	CLAMP	BACC10DU350Y		1
(d)	13	RING, WEAR	69-26229-2		1
	14	SHIM, LAMINATED	69-56508-3		1
	15	SHIM, SOLID	69-56508-6		1
(e)		NO PARTS REQUIRED			

Cabin Air Manifold
Figure 508 (Sheet 2)

(2) Assemble components of the engine starter manifold as follows:
(Fig. 509)

(a) Attach manifold assembly (1) to duct assembly (2) with wear ring (3) and shims (4 and 5) as follows:

1) Install wear ring (3) in bearing nut of duct assembly (2), and measure gap between ends of ring. Gap must not exceed 0.030 inch.

NOTE: Ends of wear ring may be filed if necessary to facilitate installation.

2) Join duct assembly (2) and manifold assembly (1) with wear ring (3) in place, but without shims. Tighten bearing nut on duct assembly (2) to 70-90 pound-inches.

3) Measure gap between bearing nut and retainer rim with feeler gage. (See gap X in detail A.)

4) Add 0.012 to 0.014 inch to gap dimension to obtain total combined thickness of shims (4 and 5) to be used for final assembly of manifold and duct. Adjust thickness of shims by removing laminations from shim (4).

NOTE: Deleted.

5) Detach duct and manifold assemblies.

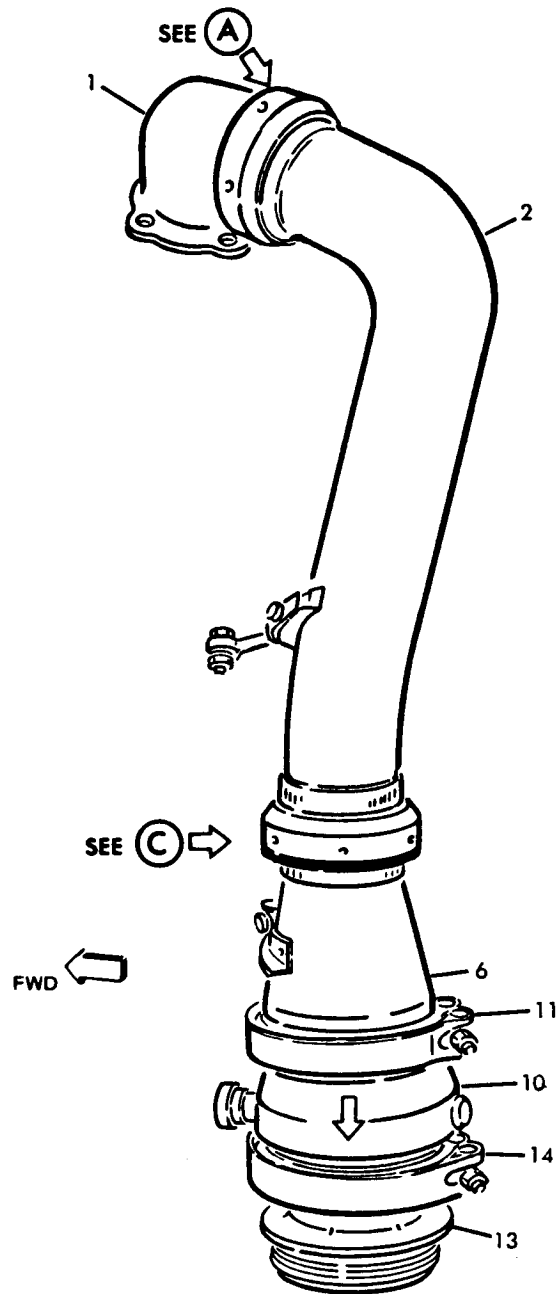
6) Coat threads of bearing nut and retainer with anti-seize compound, MIL-A-907. Reassemble duct and manifold assemblies with wear ring (3) and shims (4 and 5) in place.

NOTE: Edges of shims must be flush with outer surface of bearing nut.

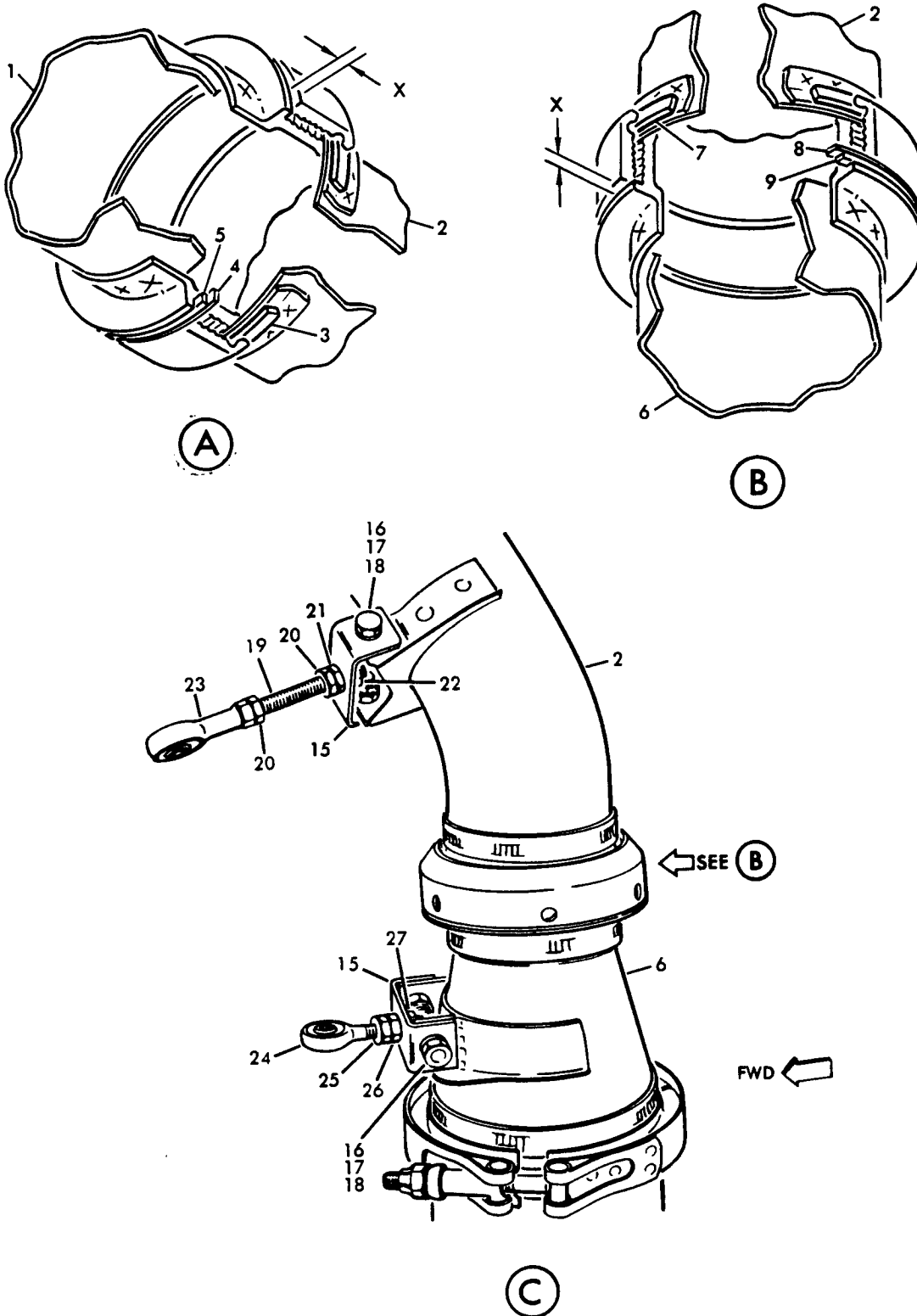
7) Tighten bearing nut to 380-400 pound-inches.

8) Check joint for freedom of movement. Lockwire bearing nut to retainer per 20-50-02.

- (b) Attach duct assembly (6) with wear ring (7) and shims (8 and 9) per step (a). (See detail B.)
- (c) Attach check valve (10) to duct assembly (6) with clamp (11). Orient valve as shown in Fig. 509.
- (d) Install retainer (13) on check valve (10) with clamp (14).
- (e) Install brackets (15) on welded brackets on duct assemblies (2 and 6) with bolts (16), bushings (18), and nuts (17).
- (f) Install rod (19) and rod end (23) on bracket (15) on duct assembly (2) with nuts (20, 21, and 22). Do not tighten nuts.
- (g) Install rod end (24) on bracket (15) on duct assembly (6) with nuts (25, 26, and 27). Do not tighten nuts.
- (h) Cap open ends of ducts.



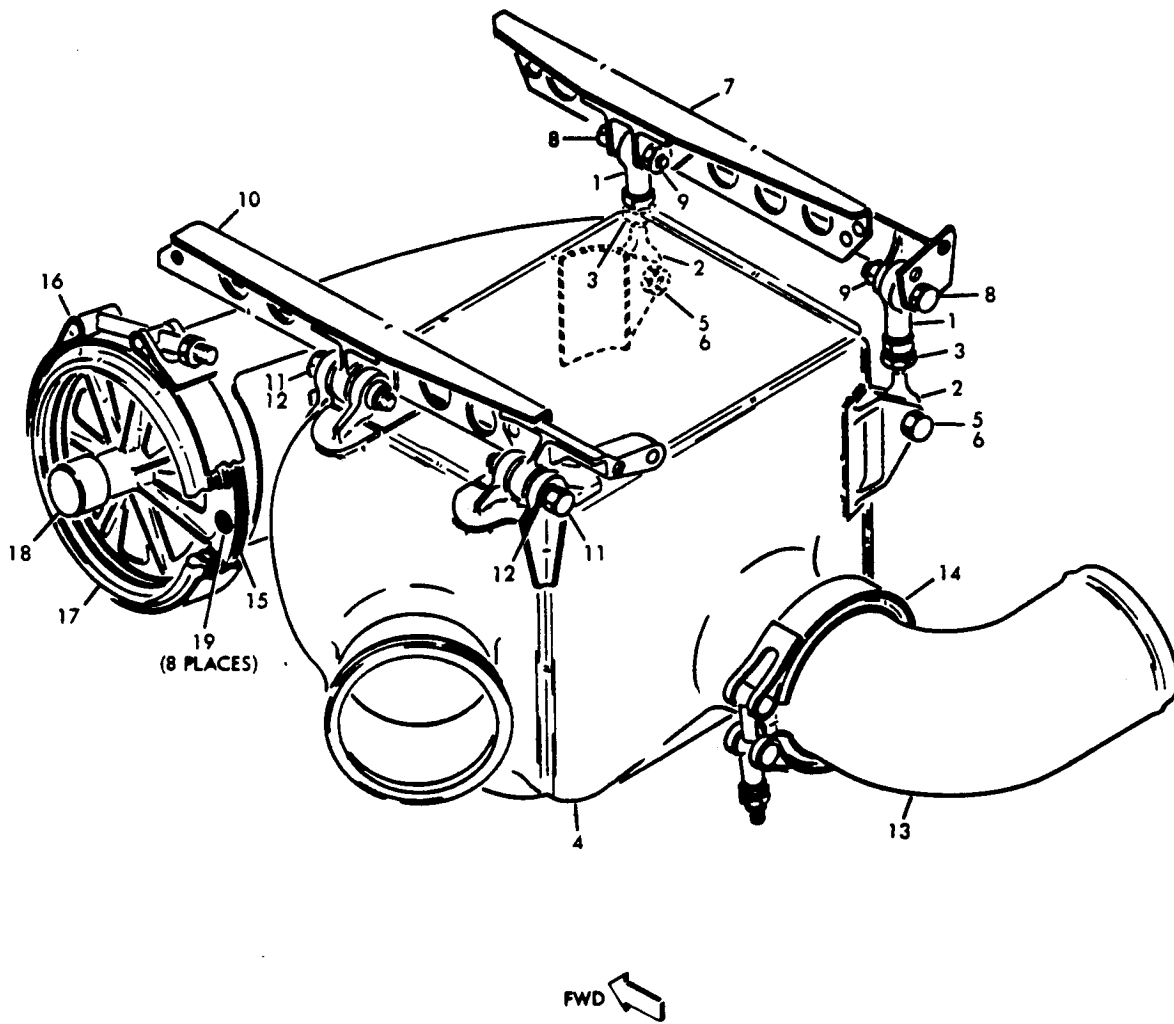
REF.
FIG
508
547
1112



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(2)(a)	1	MANIFOLD ASSEMBLY	69-45175-1		1
	2	DUCT ASSEMBLY	69-44635-1		1
	3	RING, WEAR	69-26231-2		1
	4	SHIM, LAMINATED	69-56508-1		1
	5	SHIM, SOLID	69-56508-4		1
(b)	6	DUCT ASSEMBLY	69-44634-1		1
	7	RING, WEAR	69-26231-2		1
	8	SHIM, LAMINATED	69-56508-1		1
	9	SHIM, SOLID	69-56508-4		1
(c)	10	VALVE, CHECK	123558-1-3		1
	11	CLAMP	BACC10DU350Y		1
(d)	13	RETAINER	69-17969-3		1
	14	CLAMP	BACC10DU350Y		1
(e)	15	BRACKET	69-44633-3		2
	16	BOLT	NAS1103-3		4
	17	NUT	BACN10JC3		4
	18	BUSHING	BACB28X3C010		4
(f)	19	ROD	NAS1454C4-0208		1
	20	NUT	AN316-4R		2
	21	NUT	AN315-4R		1
	22	NUT	BACN10JC4		1
	23	ROD END	BACB10J3		1
(g)	24	ROD END	BACB10J12		1
	25	NUT	AN316-4R		1
	26	NUT	AN315-4R		1
	27	NUT	BACN10JC4		1
(h)		NO PARTS REQUIRED			

Engine Starter Manifold
Figure 509 (Sheet 3)

- I. Assemble precooler with components of the air conditioning system as follows: (See figure 510.)
- (1) Assemble two rod end bearings (1) with two rod end bearings (2) and jamnuts (3). Adjust length of bearing assemblies to 2.47 inches and tighten jamnuts.
 - (2) Attach two bearing assemblies fabricated in step (1) to precooler assembly (4) with bolts (5) and nuts (6).
 - (3) Attach free ends of bearing assemblies to clevis fittings on precooler support assembly (7) with bolts (8) and nuts (9).
 - (4) Install support assembly (10) on precooler with bolts (11) and washers (12). Install washers with countersunk side against bolt.
 - (5) Attach duct (13) to aft side of precooler with clamp (14).
 - (6) Install gasket (15), lockring assembly (16), seal assembly (17), and regulator assembly (18), with eight screws (19).
 - (7) Cap or cover all open ports in assembly.



REF.
FIG
527
1113

Air-Conditioning Precooler Buildup
Figure 510 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
I. (1)	1	BEARING, Rod End	BACB10A180D		2
	2	BEARING, Rod End	BACB10A187		2
	3	JAMNUT	AN316C5		2
(2)	4	PRECOOLER ASSEMBLY	182400-1-1		1
	5	BOLT	NAS1304-9		2
	6	NUT	NAS679C4W		2
(3)	7	SUPPORT ASSEMBLY	69-42571-1		1
	8	BOLT	NAS1104-11		2
	9	NUT	NAS679C4W		2
(4)	10	SUPPORT ASSEMBLY	69-42570-1		1
	11	BOLT	69-42825-1		2
	12	WASHER	MS20002C4		2
(5)	13	DUCT	65-53867-2		1
	14	CLAMP	BACC10DU350Y		1
(6)	15	GASKET	69-42533-1		1
	16	LOCKRING ASSEMBLY	69-46653-1		1
	17	SEAL ASSEMBLY	D58364 or 69-46654-1		1
	18	REGULATOR ASSEMBLY	392664-1-1		1
(7)	19	SCREW	BACB30LU2-1		8
		No parts required			

737
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

| J. Deleted

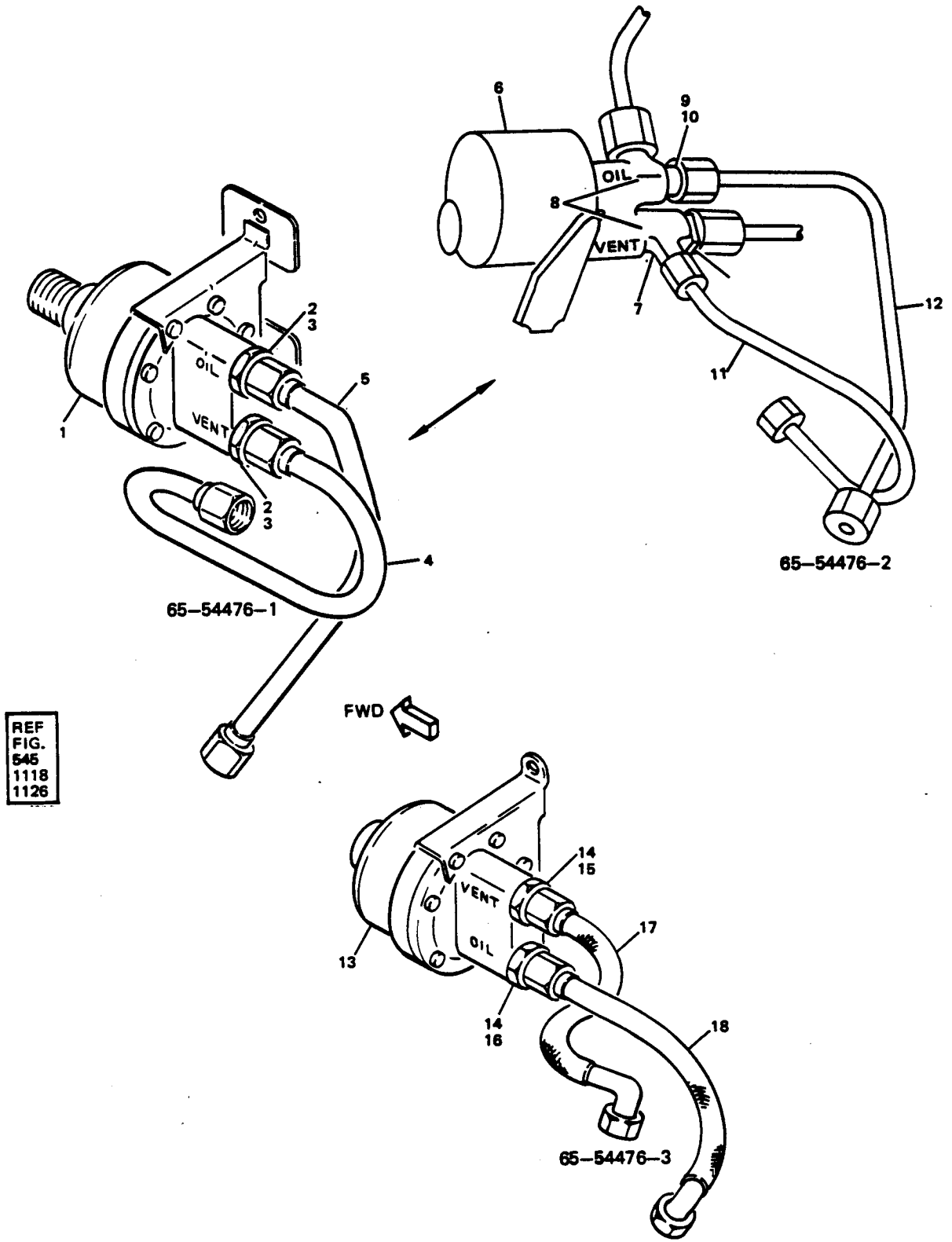
| Nov 15/68

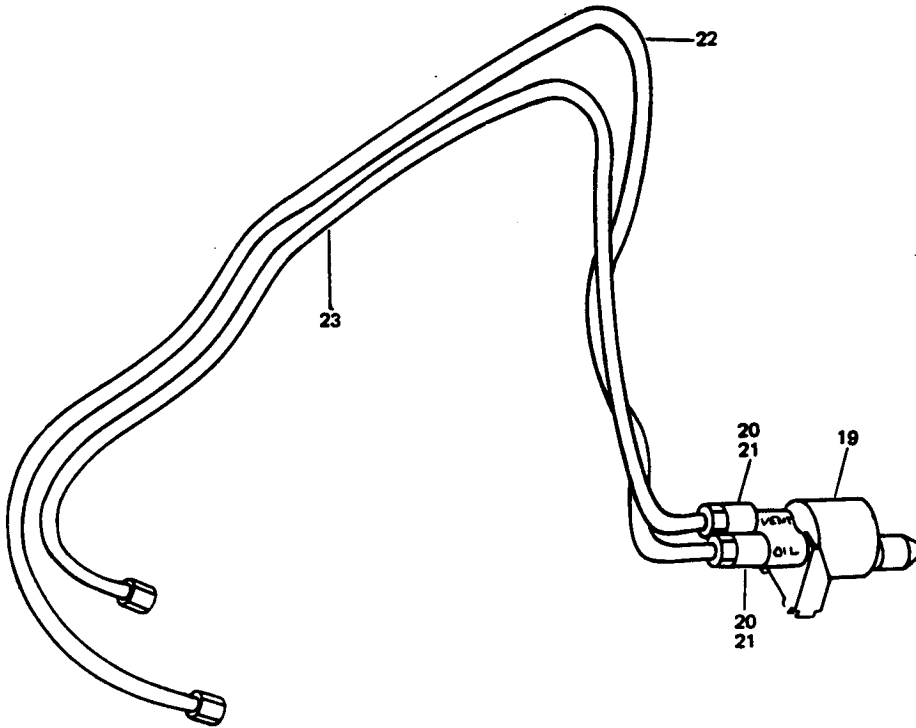
--PAGES 516D and 517 DELETED--

71-03-01
Page 516C

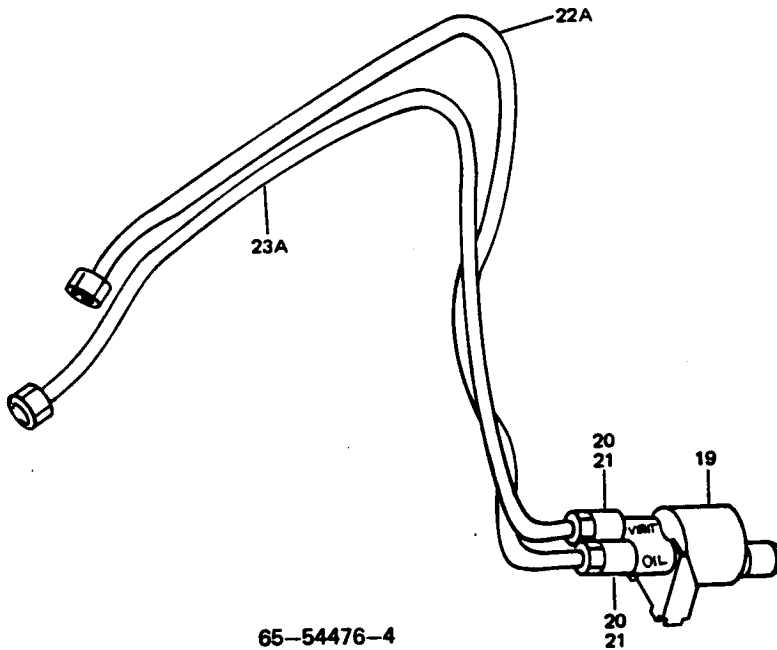
K. Assemble engine oil differential pressure switch with components of pressure sensing system as follows: (Fig. 512)

- (1) 65-54476-1 -- Install reducer union (2) with O-rings (3) and tighten. Attach tube assembly (4) to VENT port and tube assembly (5) to OIL port. Do not tighten end fittings on tube assemblies, this will be accomplished during differential pressure switch installation (Fig. 545). Cap or plug all open ports.
- (2) 65-54476-2 -- Install elbow (7) on differential pressure switch (6) using O-rings (8 and 9), and bolts (10). Attach tube assembly (11) to VENT port and tube assembly (12) to OIL port. Do not tighten end fittings on tube assemblies, this will be accomplished during differential pressure switch installation (Fig. 545). Cap or plug all open ports.
- (3) 65-54476-3 -- Install union (16) with O-ring (14) to OIL port, and adapter (15) with O-ring (14) to VENT port on differential pressure switch assembly (13). Install hose assembly (17) on adapter (15) and hose assembly (18) on union (16). Cap or plug all open ports.
- (4) 65-54476-4 -- Install reducer union (21) with O-rings (20) and tighten. Install tube assembly (22 or 22A) to VENT port and tube assembly (23 or 23A) to OIL port on differential pressure switch (19). Do not tighten end fittings on tube assemblies, this will be accomplished during differential pressure switch installation (Fig. 545). Cap or plug all open ports.





65-54476-4

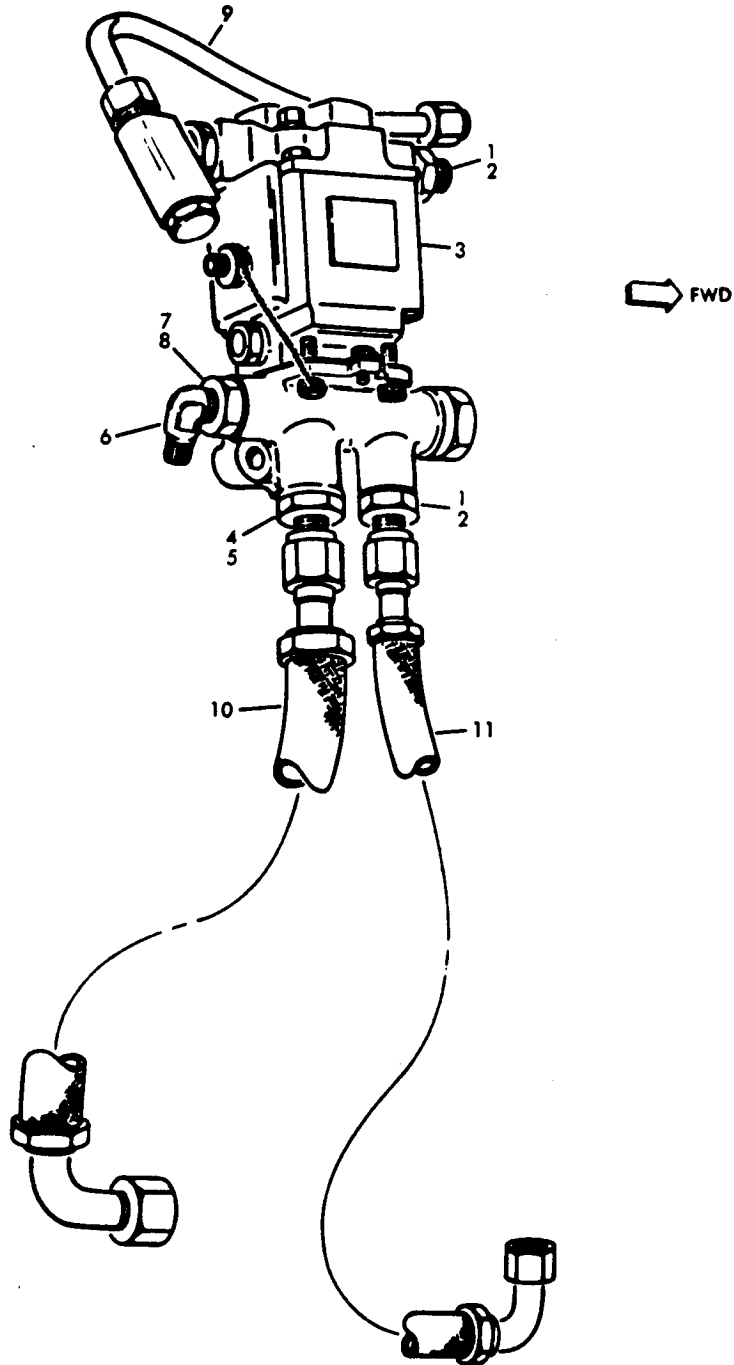


65-54476-4

Engine Oil Differential Pressure Switch
Figure 512 (Sheet 2)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
K. (1)	1	SWITCH, PRESSURE	10-3269-8		1
	2	UNION, REDUCER	MS21916-6-4		2
	3	O-RING	BACP11PB4		2
	4	TUBE ASSY	65-58135-15		1
	5	TUBE ASSY	65-58135-16		1
(2)	6	SWITCH, PRESSURE	10-3269-8		1
	7	ELBOW - 75 DEGREE	66-19214-1		2
	8	O-RING	BACP11PB4		2
	9	O-RING	BACP11P11		2
	10	BOLT	69-45458-1		2
	11	TUBE ASSY	65-58135-34		1
	12	TUBE ASSY	65-58135-35		1
(3)	13	SWITCH ASSY, PRESSURE	69-72688-1		1
	14	O-RING	BACP11PB4		2
	15	ADAPTER	MS21900-4		1
	16	UNION	MS21902-4		1
	17	HOSE ASSY	AE3660121E0066		1
	17	HOSE ASSY (OPT)	SR50206CC4-0066		1
	18	HOSE ASSY	AE3660131E0082		1
	18	HOSE ASSY (OPT)	SR50306CC4-0082		1
(4)	19	SWITCH, PRESSURE	10-3269-8		1
	20	O-RING	BACP11PB4		2
	21	UNION, REDUCER	MS21916-6-4		2
	22	TUBE ASSY	65-58135-39		1
	22A	TUBE ASSY	65-58135-42		1
	23	TUBE ASSY	65-58135-38		1
	23A	TUBE ASSY	65-58135-45		1

- I L. Build up differential pressure regulator as follows: (See figure 513.)
- (1) Install unions (1) and gaskets (2) in pressure differential regulator (3). Tighten unions.
 - (2) Install union (4) and gasket (5). Tighten unions.
 - (3) Install elbow (6) with jamnut (7) and gasket (8). Do not tighten jamnut.
 - (4) Install tube assembly (9). Do not tighten end fitting of tube assembly.
 - (5) Attach hose (10) to union (4). Do not tighten hose end fitting.
 - (6) Attach hose (11) to union (1) in bottom port of regulator. Do not tighten hose end fitting.
 - (7) Cap or plug all open ports in assembly.



REF.
FIG
529
1111

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
L. (1)	1	UNION	MS21902-6		2
	2	GASKET	BACP11PB6		2
	3	REGULATOR, PRESSURE DIFFERENTIAL	392686-3-1		1
(2)	4	UNION	MS21902-8		1
	5	GASKET	BACP11PB8		1
(3)	6	ELBOW	MS21926-4		1
	7	JAMNUT	AN924-4		1
	8	GASKET	BACP11PB4		1
(4)	9	TUBE ASSEMBLY	65-58126-11		
(5)	10	HOSE	16230 (AVICA)		1
	10	HOSE	AS138-08-0134		1
(6)	11	HOSE	16229 (AVICA)		1
	11	HOSE	AS136-06-0260		1
(7)		NO PARTS REQUIRED			

Differential Pressure Regulator
Figure 513 (Sheet 2)

M. Build up hydraulic pump as follows: (Abex/Kellogg)

NOTE: Lubricate all O-rings with Hydraulic Fluid, BMS 3-11, or Skydrol Assembly Lube MCS 352, before installation.

(1) Build up of hydraulic pump (1, Fig. 514) (1, Fig. 514A)

NOTE: See (2), below, for buildup incorporating SB 29-1024.

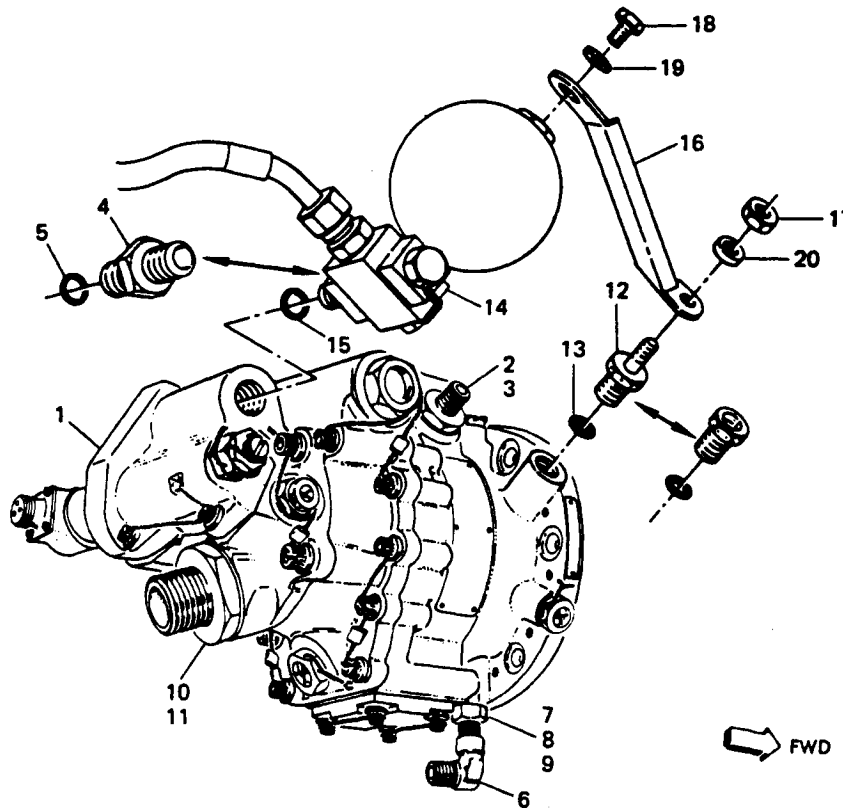
- (a) Install union (2) in pump (1) with O-ring (3).
- (b) Install reducer (4) with O-ring (5).
- (c) Thread jamnut (7) on reducing elbow (6). Install reducing elbow in pump with O-ring (8) and gasket (9).
- (d) Install reducer (10) with O-ring (11).
- (e) Cap or plug all open ports in assembly.

(2) Buildup of pump having SB 29-1024 incorporated (1, Fig. 514):

- (a) Install union (2) with O-ring (3).
- (b) Install plug (12) with O-ring (13) in place of drain plug and packing. Tighten plug 100-125 pound-inches.
- (c) Assemble reducer (4) with O-ring (5) on attenuator (14).

NOTE: An attenuator that was removed from a failed pump must be flushed with hydraulic fluid BMS 3-11 before installation.

- (d) Install attenuator on pump with O-ring (15).
- (e) Install bracket (16), using nut (17), bolt (18) and washers (19 and 20). Lockwire.
- (f) Perform steps (c), (d) and (e) in (1), above.



REF.
FIG
530
1130

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
M.					
(1)(a)	1	PUMP, Hydraulic	65001		1
	1	PUMP, Hydraulic	55098		1
	1	PUMP, Hydraulic	55097		1
	2	UNION	MS21902-6		1
	3	O-RING	NAS1612-6		1
(b)	4	REDUCER	MS21916-10-8		1
	5	O-RING	NAS1612-10		1
(c)	6	ELBOW, Reducing	BACE21AS0406		1
	7	JAMNUT	AN6289-4		1
	8	O-RING	NAS1612-4		1
	9	GASKET	MS28777-4		1
(d)	10	REDUCER	AN919-26D		1
	11	O-RING	NAS1612-20		1
(e)		No parts required			

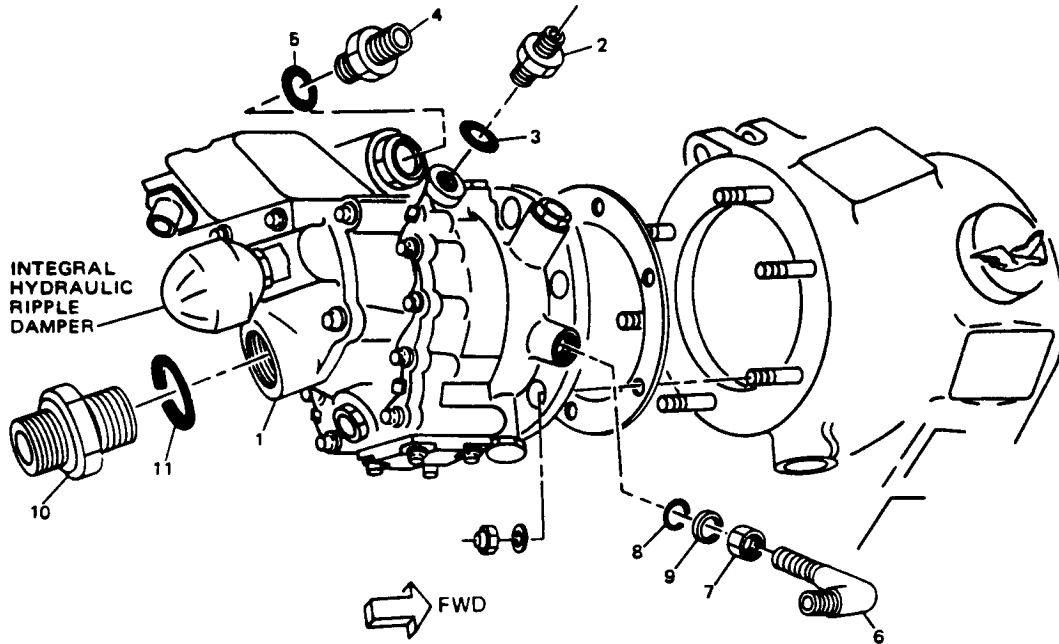
BOEING
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(2)(a)	2	UNION	MS21902-6		1
	3	O-RING	NAS1612-6		1
(b)	12	PLUG	66-12113-1		1
	13	O-RING	NAS1612-4		1
(c)	4	REDUCER	MS21916-10-8		1
	5	O-RING	NAS1612-10		1
	14	ATTENJATOR	9011520100-1		1
(d)	14	ATTENUATOR	9011520100-1		1
	15	O-RING	NAS1612-10		1
(e)	16	BRACKET *[1]	69-65327-2 or -3		1
	16	BRACKET *[1]	69-67501-2 or -3		1
	17	NUT	BACN10JC4		1
	18	BOLT	BACB30NE5H1		1
	19	WASHER	AN960-516		1
	20	WASHER	AN960-416		1

*[1] BRACKET 69-65327-2 or 69-67501-2 used with pump 55097 and 65001
 BRACKET 69-65327-3 or 69-67501-3 used with pump 55098

Hydraulic Pump Buildup (Abex/Kellogg)
 Figure 514 (Sheet 2)



REF
FIG.
530
1130A

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
M. (1)	(a) 1	PUMP, Hydraulic	55098-01		1	
	2	UNION	MS21902-6		1	
	3	O-RING	NAS1612-6		1	
	(b) 4	REDUCER	MS21916-10-8		1	
	5	O-RING	NAS1612-10		1	
	(c) 6	ELBOW, Reducing	BACE21AS0406		1	
	7	JAMNUT	AN6289-4		1	
	8	O-RING	NAS1612-4		1	
	9	GASKET	MS28777-4		1	
	(d) 10	REDUCER	AN919-26D		1	
	11	O-RING	NAS1612-20		1	
	(e)		NO PARTS REQUIRED			

- (3) Build up Vickers yoke type hydraulic pump (Fig. 515).

NOTE: Apply hydraulic assembly lubricant MCS 352 to O-rings before installation.

- (a) Install union (2) in hydraulic pump (1) with O-ring (3).
- (b) Install reducer (4) with O-ring (5).
- (c) Thread jamnut (7) on reducing elbow (6). Install reducing elbow in pump with O-ring (8) and gasket (9).
- (d) Install union (10) with O-ring (11).
- (e) Cap or plug all open ports in assembly.

- (4) Build up Vickers in-line type hydraulic pump (Fig. 515A).

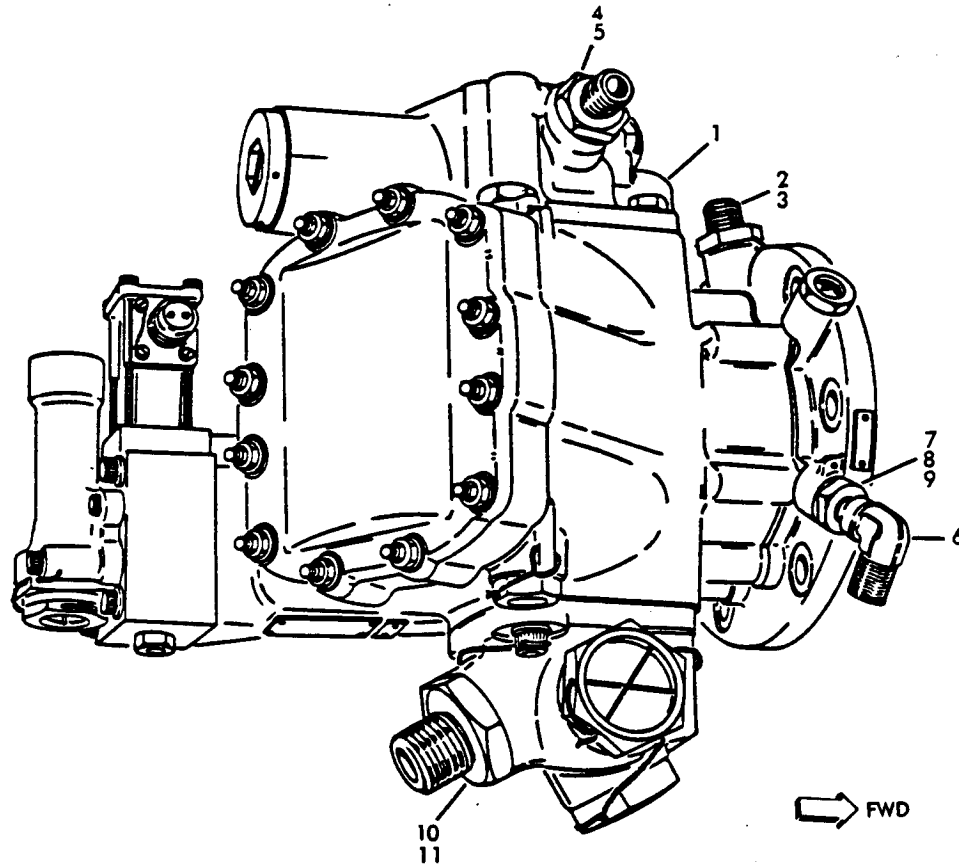
NOTE: See (5) below for buildup incorporating SB 29-1024.

Apply hydraulic assembly lubricant MCS 352 to O-rings before installation.

- (a) Install union (2) with O-ring (3).
- (b) Install reducer (4) with O-ring (5).
- (c) Install union (6) with O-ring (7).
- (d) Install elbow (8), nut (9), O-ring (10) and back-up ring (11), with elbow outlet facing aft.
- (e) Install dust caps or plugs on open ports.

- (5) Build up Vickers in-line type pump having SB 29-1024 incorporated.

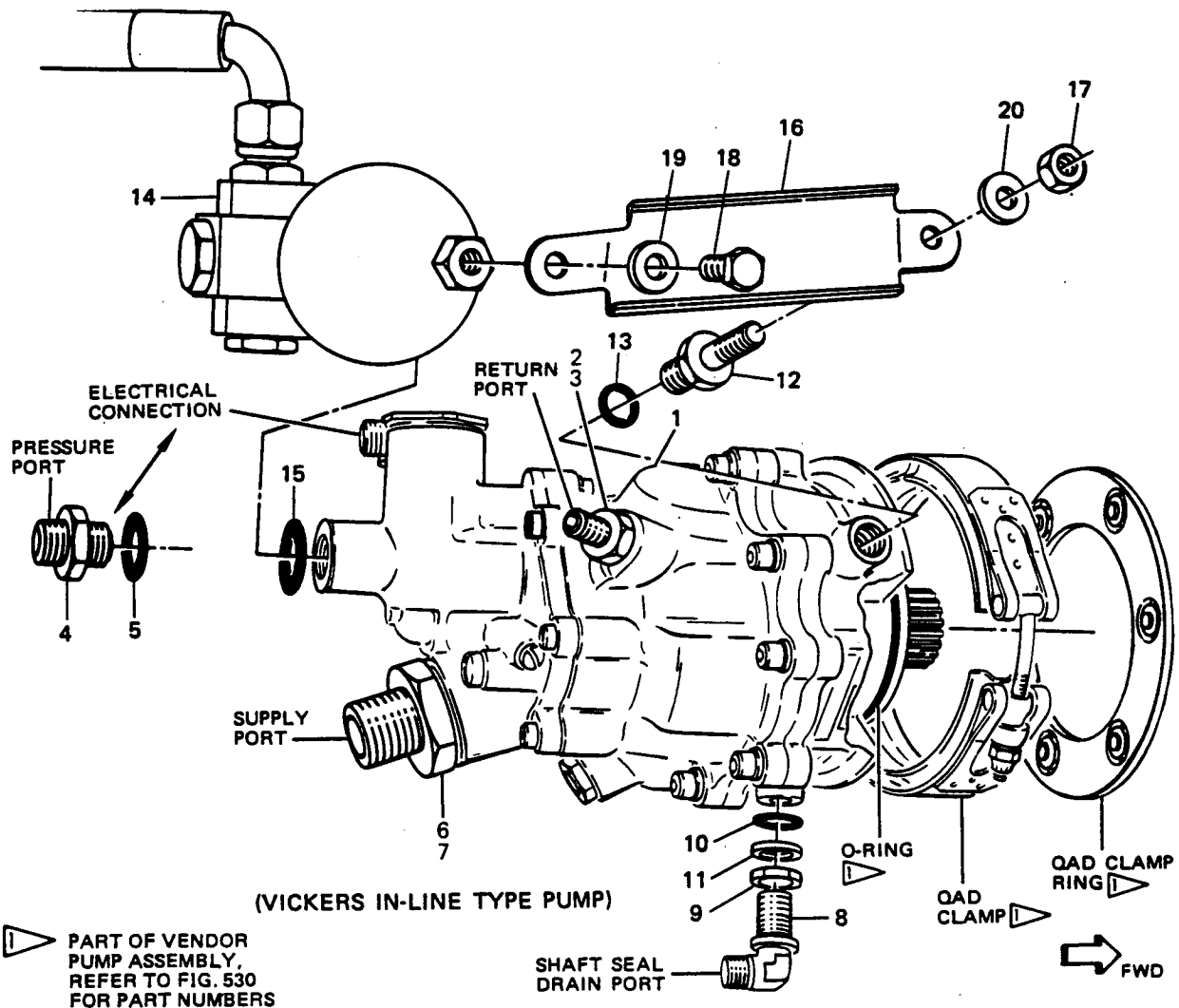
- (a) Install union (2) with O-ring (3).
- (b) Install plug (12) with O-ring (13) in place of drain plug and packing. Tighten plug 100-125 pound-inches. Lockwire.
- (c) Assemble reducer (4) with O-ring (5) on attenuator (14).
- (d) Install attenuator on pump with O-ring (15).
- (e) Install bracket (16), using nut (17), bolt (18) and washers (19 and 20). Lockwire.
- (f) Perform steps (c), (d) and (e) in (1), above.



REF.
FIG
530
1130

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
M. (3)(a)	1	PUMP, Hydraulic (yoke type)	10-60246-8		1
	1	PUMP, Hydraulic (yoke type)	10-60246-10		1
	2	UNION	MS21924-6		1
	3	O-RING	NAS1612-6		1
(b)	4	REDUCER	MS21916-10-8		1
	5	O-RING	NAS1612-10		1
(c)	6	ELBOW, Reducing	BACE21AS0406		1
	7	JAMNUT	AN6289-4		1
	8	O-RING	NAS1612-4		1
	9	GASKET	MS28777-4		1
(d)	10	UNION	AN815-16D		1
	11	O-RING	NAS1612-16		1
(e)		No parts required			

Hydraulic Pump Buildup (Vickers Yoke Type)
Figure 515



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
M.					
(4)(a)	1	PUMP, Hydraulic (in-line)	10-61794-1		1
	2	UNION	MS21902-6		1
	3	O-RING	NAS1612-6		1
(b)	4	REDUCER	MS21916-10-8		1
	5	O-RING	NAS1612-10		1
(c)	6	UNION	AN919-26D		1
	7	O-RING	NAS1612-20		1
(d)	8	ELBOW	BACE21AS0406		1
	9	NUT	AN6289-4		1
	10	O-RING	NAS1612-4		1
	11	BACK-UP RING	MS28777-4		1

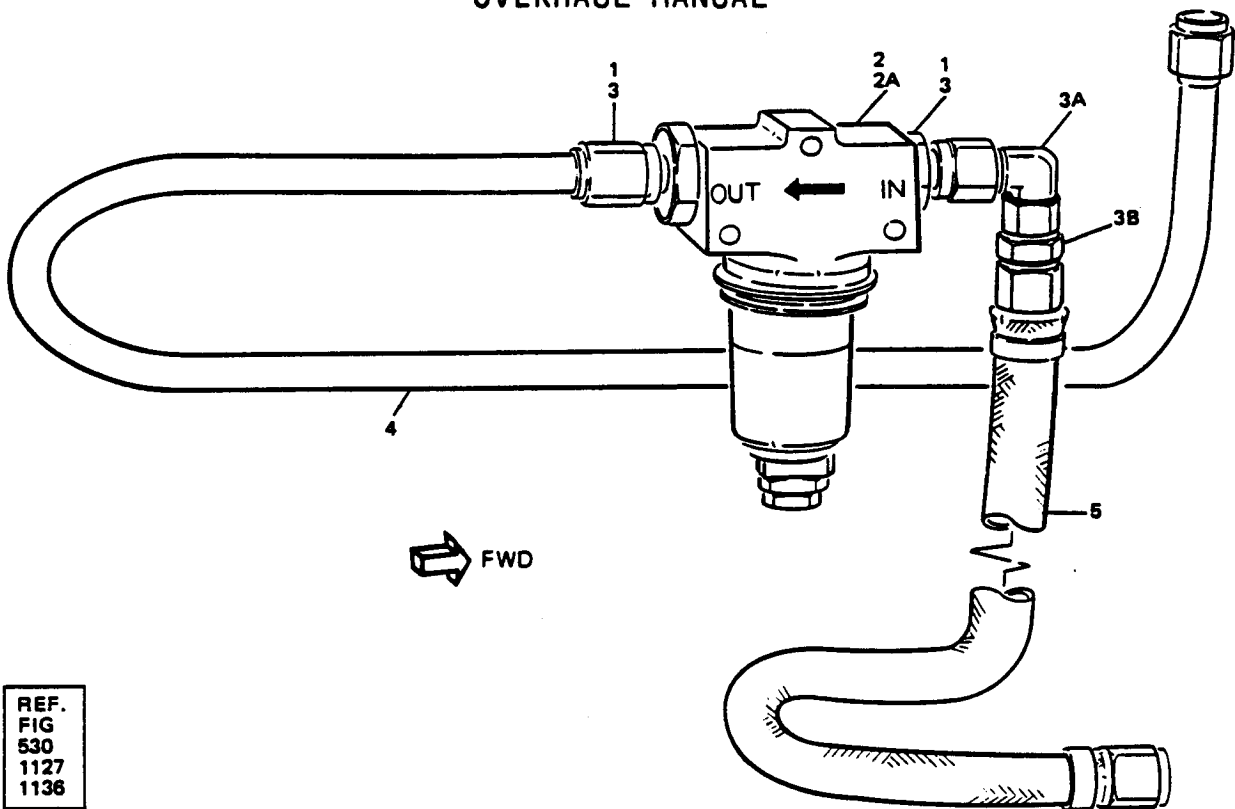
Hydraulic Pump Buildup (Vickers In-line Type)
Figure 515A (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
M. (5)	(a) 2	UNION	MS21902-6		1	
	3	O-RING	NAS1612-6		1	
	(b)	12	PLUG	66-12113-1		1
		13	O-RING	NAS1612-4		1
	(c)	4	REDUCER	MS21916-10-8		1
		5	O-RING	NAS1612-10		1
		14	ATTENUATOR	9011520100		1
	(d)	14	ATTENUATOR	9011520100		1
		15	O-RING	NAS1612-10		1
	(e)	16	BRACKET	69-65327-1		1
		16	BRACKET	69-67501-1		1
		17	NUT	BACN10JC4		1
		18	BOLT	BACB3ONE5H1		1
		19	WASHER	AN960-516		1
		20	WASHER	AN960-416		1

N. Build up hydraulic filter as follows (Fig. 516):

NOTE: Lubricate all O-rings with Skydrol Assembly Lube MCS 352, or Hydraulic Fluid BMS 3-11, before installation.

- (1) Install unions (1) in ports of filter (2, 2A) with O-rings (3). Tighten unions. Install elbow (3A) and union (3B) at inlet port.
- (2) Attach tube assembly (4) to union (1) in outlet port of filter (2). Do not tighten end fitting of tube assembly.
- (3) Attach hose (5) to union (3B) in inlet port of filter (2). Do not tighten end fitting of hose.
- (4) Cap open ends of tube and hose.



REF.
FIG
530
1127
1136

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(1)		DELETED			
N. (2)	1	UNION	MS21902-6		2
	2	FILTER	10-60555-3		1
	2A	FILTER	10-60555-11		1
	3	O-RING	NAS1612-6		2
	3A	ELBOW	BACE21AW606		1
	3B	UNION	LER1832-6, V11328 (PREF)		1
	3B	UNION	ER1832-6, V79470 (OPT)		1
(3)	4	TUBE ASSEMBLY	65-58145-7 (PREF)		1
	4	TUBE ASSEMBLY	65-58145-5 (OPT)		1
(4)	5	HOSE	10-60563-16 (USED ON 65-58145-1)		1
	5	HOSE	10-60563-29 (USED ON 65-58145-4, -9, -10, -12, -14, -15) (PREF)		1
	5	HOSE	10-60563-21 (USED ON 65-58145-4, -9, -10, -12, -14, -15) (OPT)		1
(5)		NO PARTS REQUIRED			

Hydraulic Filter Buildup
Figure 516

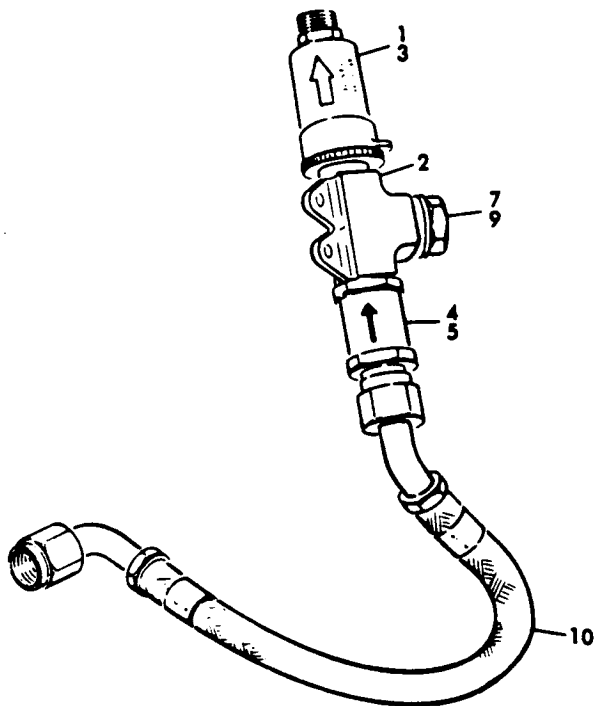
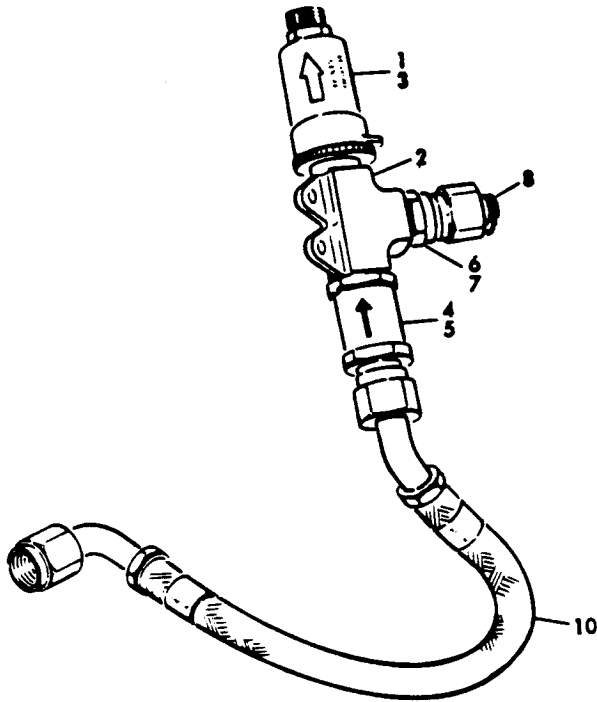
O. Preassemble components of nose cowl thermal anti-icing system into two subassemblies as follows (figures 517 and 518):

(1) Assemble TAI filter and valve as follows (figure 517):

- (a) Attach filter assembly (1) to tee (2) with gasket (3).
- (b) Install gasket (5) and check valve (4) on end of tee (2) with directional arrow pointing toward tee.
- (c) Install gasket (7) and either ground source connector (6) and cap (8) or, if specified, plug (9) inside port of tee (2).

NOTE: When installing ground source connector (6), make certain that arrow points toward tee (2).

- (d) Lockwire plug (9), if installed at step (c), per Subject 20-50-02.
- (e) Attach hose (10) to check valve (4). Do not tighten end fitting of hose.
- (f) Cap all open ports in assembly.



STEPS A THROUGH F

REF.
FIG
534
1124

Nose Cowl TAI Filter and Valve
Figure 517 (Sheet 1)

Feb 15/69

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
0.					
(1)(a)	1	FILTER ASSEMBLY	66-8324-1		1
	2	TEE	69-44477-1		1
	3	GASKET	AN901-8C		1
(b)	4	VALVE, Check	8C150		1
	5	GASKET	AN901-8C		1
(c)	6	CONNECTOR, Ground Source	411-00117(Opt)		1
	6	CONNECTOR, Ground Source	1112-58997-1		1
	7	GASKET	AN901-8C		1
	8	CAP	BACC14H8LS		1
	9	PLUG	BACP20AU-8		1
(d)		No parts required			
(e)	10	HOSE	16225		1
(f)		No parts required			
		Steps (g) through (m) deleted			

 Jan 5/77
 +

 Nose Cowl TAI Filter and Valve
 Figure 517 (Sheet 2)

 71-03-01
 Page 522C
 (Page 522D BLANK)

(2) Assemble components of nose cowl TAI system with thermostatic valve as follows: (See figure 518.)

(a) Position gasket (1) on shutoff valve (2) and attach thermostatic valve (3) with two bolts (4), washers (5), and nuts (6) in positions visible in figure 518. In opposite side of flange, loosely install two bolts (7), washers (5), and nuts (6) to be used in attaching valve assembly to engine structure.

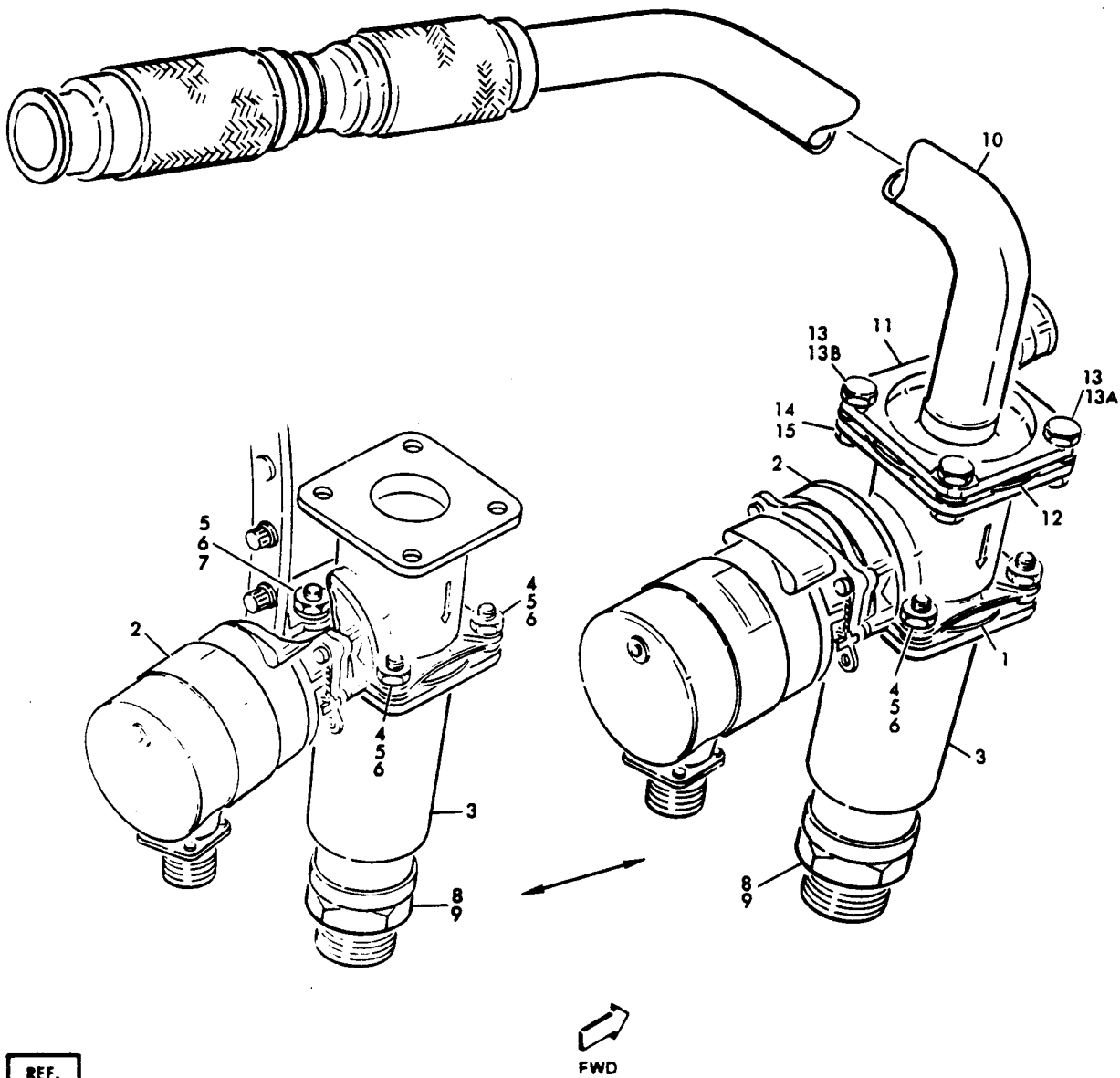
(b) Install gasket (8) and reducer (9) on end of thermostatic valve. If bracket 65-45799-129 or -141 is used in mounting TAI valve on engine flange D, use two gaskets (8) and do not tighten reducer (9).

NOTE: Steps (c) through (d) apply only for engines without vortex dissipator. Step (e) applies to all installations.

(c) Insert end of tube assembly (10) through flange (11).

(d) Seat flange on duct assembly (10) in flange (11). Place gasket (12) in position and attach flange (11) to shutoff valve (2) with four bolts (13) or two bolts (13A) and two bolts (13B) with washers (14), and nuts (15).

(e) Cap all open ports in assembly.



REF.
FIG
534
1124
1125

Nose Cowl TAI Thermostatic Valve
Figure 518 (Sheet 1)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
0.					
(2)(a)	1	GASKET	60024 *[1]		1
	1	GASKET	60023 *[1]		1
	2	VALVE, Shutoff	10-60817-8 *[5]		1
	2	VALVE, Shutoff	10-60817-7		1
	2	VALVE, Shutoff	10-60817-10 *[5]		1
	3	VALVE, Thermostatic	392708-1-1 *[3]		1
	3	VALVE, Thermostatic	10-60792-1 *[3]		1
	4	BOLT	BACB30LM5U9		2
	4	BOLT	AN5C10A (opt)		2
	5	WASHER	AN960C516		4
	6	NUT	NAS679C5		4
	7	BOLT	AN5C11A (opt)		2
	7	BOLT	BACB30LM5U11 (used with 10-60817-8)		2
(b)	8	GASKET	AN901-24C		2 or 1
	9	REDUCER	AN919-28J		1
(c)	10	TUBE ASSY	65-58123-2		1
	11	FLANGE	BACF22S32AC		1
(d)	12	GASKET	60023		1
	12	GASKET	60023-1 *[4]		1
	13	BOLT	AN5C10A *[2]		4
	13A	BOLT	BACB30LM5U10		2
	13B	BOLT	BACB30LM5U12		2
	14	WASHER	AN960C516		4
	15	NUT	NAS679C5		4
(e)		No parts required			

*[1] 60023 used with 10-60792-1; 60024 used with 392708-1-1

*[2] Optional to BACB30LM5U10 with BACB30LM5U12

*[3] 392708-1-1 replaces 10-60792-1

*[4] 60023-1 replaces 60023

*[5] 10-60817-10 optional to 10-60817-8

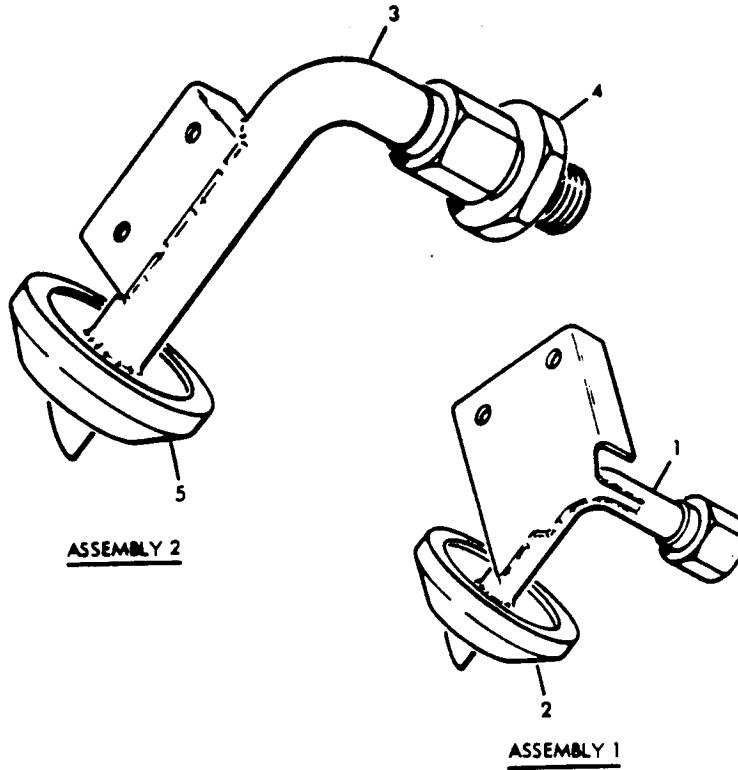
P. Preassemble components of engine drain system into two drain fitting assemblies as follows: (See figure 519.)

(1) Drain fitting assembly 1:

- (a) Install seal (2) on engine drain fitting assembly (1).
- (b) Cap open ends of assembly.

(2) Drain fitting assembly 2:

- (a) Install union (4) in engine drain fitting assembly (3).
- (b) Install seal (5) on engine drain fitting assembly (3).
- (c) Cap open ends of assembly.



REF.
FIG
551
1116

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
P. (1)(a)	1	FITTING ASSEMBLY, Engine Drain	69-39486-4		1
	2	SEAL	69-39494-2		1
(b)		NO PARTS REQUIRED			
(2)(a)	3	FITTING ASSEMBLY, Engine Drain	69-62277-1		1
	3	FITTING ASSEMBLY, Engine Drain	69-39487-1		1
	4	UNION	MS21902-12		1
(b)	5	SEAL	69-39494-1		1
(c)		No parts required			

Engine Drain Fitting Assemblies
Figure 519

4. Power Plant Buildup Sequence

- A. Installation instructions have been arranged so that components can be installed on the engine in a logical sequence, moving from one area to an adjacent area, to preclude unnecessary, haphazard skipping back and forth across the engine. Recommended sequence of installation is in the same order as the component installation index.

INSTALLATION	PARAGRAPH	FIGURE
Engine Accessory Support Bracket	5.A.	520
Cone Bolt	5.B.	521
CSD and AC Generator	5.C.	522
AC Generator Cooling Ducts and Hook Assembly	5.D.	523
Pressure Ratio Plumbing	5.E.	524
Electrical	5.F.	525
Fire Detector	5.G.	526
Air Conditioning Precooler	5.H.	527
8th and 13th Stage Air Conditioning Manifolds	5.I.	528
Pressure Differential Regulator	5.J.	529
Hydraulic System Components	5.K.	530
N2 Tachometer Generator	5.L.	531
Strut Drain Tubing	5.M.	532
Air Conditioning Sensing Plumbing	5.N.	533
Nose Cowl Anti-Icing System Components	5.O.	534
Fuel Inlet Tube	5.P.	535
Engine Controls Rigging Pin Mount	5.Q.	536

INSTALLATION	PARAGRAPH	FIGURE
Fuel Heater Air Exhaust Duct	5.R.	537
Engine Drain Plumbing	5.S.	538
Fuel Flowmeter	5.T.	539
Oil Tank Accessory Components	5.U.	540
Gearbox Overboard Vent Fitting	5.V.	541
Inlet Vibration Pickup	5.W.	542
Engine Oil Pressure Transmitter and Switch	5.X.	543
Engine Oil Temperature Transmitter	5.Y.	544
Engine Oil Differential Pressure Switch	5.Z.	545
Engine Starter	5.AA.	546
Cabin Air and Engine Starter Ducts	5.AB.	547
Engine Control Linkage	5.AC.	548
CSD Oil Filter	5.AD.	549
CSD Oil Cooler	5.AE.	550
Aft Engine Drain Fittings	5.AF.	551
CSD Pressure System Plumbing	5.AG.	552
P & D Drain Tube and Combustion Chamber Fuel Drain Tube	5.AH.	553
Turbine Vibration Pickup	5.AI.	554
Hydraulic Tubes	5.AJ.	555

INSTALLATION	PARAGRAPH	FIGURE
Tail Plug	5.AK.	556
DELETED		557
Tailpipe Assembly and Shroud Assembly	5.AM.	558
N1 Tachometer Generator	5.AO.	560
Nose Dome	5.AP.	561
Chip Detector	5.AQ.	561A
Nose Cowl Installation, Engine No. 1	6.A.	562
Thrust Reverser, Hydraulic Type, Engine No. 1	6.B.	563
Fire Detector and Wire Bundle, Tailpipe Assembly, Engine No. 1	6.C.	564
Plumbing Installation, Engine No. 1	6.D.	565
Striker, Thrust Reverser Flipper Door	6.F.	566A
Nose Cowl Installation, Engine No. 2	7.A.	567
Thrust Reverser, Hydraulic Type, Engine No. 2	7.B.	568
Fire Detector and Wire Bundle, Tailpipe Assembly, Engine No. 2	7.C.	569
Plug Installation, Engine No. 2	7.D.	570
Striker, Thrust Reverser Flipper Door	7.F.	571A

5. Buildup of Engine to Maximum Neutral Configuration

A. Install support brackets for engine accessories, wire bundles, tubes, and ducts, as follows: (See figure 520.)

NOTE 1: Flange bolt holes are numbered on illustration to help locate brackets. Count down from top centerline of engine on either left or right side to find location of bracket.

NOTE 2: Attach brackets using flange bolts. Remove and reinstall flange bolts as required.

NOTE 3: At specific locations longer flange bolts are required. At these locations alternative types of bolts and nuts have been designated optional. In other positions the next size longer bolt may be substituted where necessary for existing flange bolts.

NOTE 4: After removing and reinstalling engine flange bolts to attach bracket, tighten bolts 110 to 150 pound-inches (on all flanges). Reinstall lockwire as required.

- (1) Installation of brackets on flange "B": (See view 1.)
 - (a) Install bracket (2) on aft side with flange forward.
 - (b) On engines with bracket installation 65-45799-123, install seven brackets (3), and for others install three bracket assemblies (2A) on aft side with flanges aft. Attach each bracket assembly with one flange bolt.
 - (c) On engines with vortex dissipator, install bracket (3A) at hole positions 21 and 22 on aft side with flanges aft. Remove bracket (3) at hole position 21, and reinstall on top of bracket (3A).
 - (d) On bracket (3A) install brackets (3B and 3C), securing each with two bolts (3D), nuts (3E), and washers (3F).
 - (e) On engines without vortex dissipator, install bracket (4) on aft side with flange forward, and install bracket assembly (4B) on top of bracket (4) at hole position 26 and with washer (4A) as spacer at hole position 27.
 - (f) On engines with vortex dissipator, install bracket (4B) first. Then install brackets (4 and 4C) on top with spacing washers (4A) at hole positions 25 and 28.
 - (g) Install brackets (5 and 6) together on aft side with flange aft. Attach with three flange bolts, with one bolt common to both brackets.
 - (h) Install bracket (6A) on aft side with flange aft, using center bolt of three bolts attaching P&W bracket 464464.

- (2) Installation of brackets on flange "C": (See view 2.)
- (a) Install bracket assembly (7) on forward side with flanges forward.
 - (b) Install seven bracket assemblies (8) on forward side with flange of each forward.
 - (c) Install two bracket assemblies (9) on forward side with flange of each forward.
 - (d) Install bracket (10) on aft side.
 - (e) Install bracket assembly (11) on aft surface of bracket (10) with three screws (12) and washers (13). (See detail A.)
 - (f) Install bracket assembly (14) on forward side with long flange forward and bracket assembly (14A) on aft side with flange aft in hole positions 16, 17, and 18 with one bolt common to both. Install second bracket assembly (14) in holes 22 and 23 on forward side with long flange forward.
 - (g) Install bracket assembly (15) on forward side with flange aft.
 - (h) Install bracket assembly (16) on forward side with flanges forward.
 - (i) Install bracket assembly (17) on aft side with flanges aft.
- (3) Installation of brackets on flange "D": (See view 3 and 3A.)
- (a) If required, install two bracket assemblies (18) on forward side with flanges forward.
 - (b) Install three bracket assemblies (19) on forward side with flanges forward.
 - (c) Install bracket assembly (20) on forward side with flange forward.
 - (d) Install bracket assembly (20A) on forward side with flange forward.
 - (e) Install bracket assembly (20B) on forward side with flange forward.
 - (f) Install two bracket assemblies (20C) on forward side with flange forward.
 - (g) Deleted

- (h) Deleted.
- (i) Install two bracket assemblies (23) on forward side with flange aft.
- (j) Install bracket assembly (23A) or bracket (23B) on forward side with flange forward.
- (k) Install brackets (24, 24A) on forward side with flange forward.

NOTE: Dependent on configuration, either of two brackets (24 or 24A) may be installed, or a combination of both. Refer to parts list for applicability.

- (l) Install bracket (25) on forward side, and bracket (25A) on aft side.
 - (m) Install bracket assembly (26) on forward side with flange forward.
 - (n) Install bracket (27) on forward side with flange forward. Use bolts (27A) and nuts (27B).
 - (o) Install brackets (28, 29) on forward side with flanges forward, or bracket (27C) on aft side with flanges aft.
 - (p) Deleted
 - (q) Deleted
 - (r) Deleted
 - (s) Install bracket (29D) on aft side with flange aft.
- (4) Installation of brackets on flange E: (Ref view 4)
- (a) Install bracket (30) on forward side with flange aft.
 - (b) Install six bracket assemblies (31) on forward side with flanges forward.
 - (c) Install one bracket assembly (31) and bracket assembly (32) together as shown in detail B.
 - (d) Install two bracket assemblies (33) on aft side with flanges aft.
 - (e) If used, install bracket (33A) on forward side with flange forward. Install bracket assembly (34) on aft side with flange aft, or install bracket (34A) on forward side with tall flange forward.

- (f) For 65-45799-165 instl, install parts (35A thru 35G). (Note configuration differences and direction as shown in alternate view). For all other configurations, install brackets (35, 36, 37) on aft side with flanges facing aft.
- (g) Deleted
- (h) Deleted
- (i) Install two bolts (39) in bracket (38) with retainers (40). Attach retainers with rivets (41). Countersink outer surface of bracket for heads of rivets (detail C).
- (j) Install bracket (38) and bracket (42) as shown in detail C.
- (k) Install two brackets (42) on forward side with flanges forward.
- (l) Install bolt (42C) on bracket (42A) with retainer (42B) attached with two rivets (42D). Countersink outer surface of bracket for rivet heads.
- (m) Install bracket (42A) on aft side with flange aft.
- (n) Install two bolts (44) in bracket (43) with retainers (45). Attach retainers with rivets (46). Countersink outer surface of bracket for heads of rivets.
- (o) Install brackets (43, 47) as shown in detail D.
- (p) Install bracket (48) on forward side with flange forward.
- (q) Remove two bolts and washers from FA1 valve support fitting. Locate bracket assembly (49) and reinstall bolts without washers (view 5).
- (r) Attach raceway (50) to bracket assemblies (31) and (49) with washers (51) and screws (52).
- (s) Install raceway (53). Attach to bracket assemblies (33) and (34) with two screws (52).

(5) Installation of brackets on flange F (view 6):

- (a) Install two bolts (55) in large flange of bracket assembly (54) with retainers (56). Attach retainers with screws (57) and nuts (58). Countersink outer surface of bracket for heads of screws. Rivet nutplate (58A) to bracket (54) using rivets (58B) (detail F).
- (b) Install bracket assembly (54) on forward side with large flange forward. Attach with four flange bolts.

NOTE: Make electrical bond between bracket and engine flange. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).

- (c) Install two brackets (59) on forward side with flanges forward.

NOTE: Make electrical bond between bracket and engine flange. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).

- (d) Install bracket (60) on forward side with flange aft. Make electrical bond as in step (c).
- (e) Install bracket assembly (61) on forward side with flange forward.
- (f) Install bracket (62) on forward side with flange aft. Make electrical bond as in step (c).
- (g) Install bracket assembly (63) on forward side with flange forward.

(6) Installation of brackets on flange G (view 7):

- (a) Install bracket (64) on aft side with flange forward.
- (b) Install bracket assembly (65) on aft side with flange aft.
- (c) Install two bracket assemblies (66) on aft side with flanges aft.

NOTE: Make electrical bond between bracket assemblies and flange G. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).

- (d) Install bracket assembly (67) on aft side with flange aft.
- (e) Install bracket (68) on aft side with flange aft.

- (f) Install bracket (68A) on forward side with flanges forward and (68B) on aft side with flange aft.
 - (g) Install bracket (68C) on bracket (66) using screw (68D) and washer (68E).
- (7) Installation of brackets on flange H: (See view 8.)
- (a) Install bracket assembly (69) on aft side with flange forward.
 - (b) Install bracket assembly (70) on aft side with flange forward.
 - (c) Remove existing flange bolts and install bracket assembly (71) as shown in detail G using longer bolts (72A) and nuts (72B). Bracket assembly (71) consists of bracket (71A) and bracket (72) assembled with a spacer, bolt, nut, washer, and cotter pin.
 - (d) Install bracket (73) on forward side with flange aft.
 - (e) Install bracket (74) on forward side.
- NOTE: Make electrical bond between bracket and engine flange. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).
- (f) Install bracket assembly (75) on aft side with flanges aft.
 - (g) Install bracket assembly (76) on forward side with flanges forward.
 - (h) Install bracket assembly (77) on aft side with flange aft.
 - (i) Install bracket (78) on forward side.
 - (j) Install two fittings (79), one in hole positions 71 and 72 on aft side and one in hole positions 75 and 76 on forward side. Prior to installing bolt in hole position 76, install bracket (78A) on aft side at hole positions 76 and 77. Increase position 76 bolt length as required.
 - (k) Remove existing flange bolts and install bracket assembly (80) as shown in detail H using longer bolts (81A) and nuts (81B). Bracket assembly (80) consists of bracket (80A) and bracket (81) assembled with a spacer, bolt, nut, washer, and cotter pin.
 - (l) Install bracket assembly (82) on forward side with flange forward.
 - (m) Install bracket assembly (82A) on bracket (82E), located on forward side of flange H, with screw (82B), washer (82C), and nut (82D).

(8) Brackets installed on flange J (view 9):

- (a) Install five bracket assemblies (83) on aft side with flanges aft.
- (b) Install bracket assembly (84) on aft side with flange aft.
- (c) Install bracket assembly (85) on forward side with flange aft.
- (d) Install bracket assembly (86) on forward side with flange forward.
- (e) Install bracket (87) on aft side.
- (f) Install bracket (88) and bracket assembly (89) together as shown in detail J.
- (g) Install bracket (90) on forward side.
- (h) Install bracket (91) on aft surface of bracket (90), with flanges turned aft. Attach with screws (92, 92A), nuts (93, 93A) and spacers (93B).
- (i) Install bracket assembly (94) on forward side with flange forward.
- (j) Install bracket assembly (95) and bracket (96) together as shown in detail K. Install drilled head bolt (96A) at hole position 58 with head on forward side and engaging threads in bracket (96). Lockwire bolt head to adjacent nut.
- (k) Install bracket (97) on forward side and bracket assembly (98) on aft side. Attach brackets with three flange bolts, with two bolts common to both brackets.
- (l) Install bracket assembly (99) on aft surface of bracket (97) with screws (100) and nuts (101). Install bracket (97A) on underside of bracket (97).
- (m) Install two bracket assemblies (102), one on aft side with flange forward at hole positions 64 and 65, and one on forward side with flange forward at hole positions 67 and 68. Bolt at hole 67 serves both bracket (102) and existing P&W bracket.
- (n) Install bracket (102A) on existing P&W bracket with screws (102B), washers (102D) and nuts (102C).

- (o) Install two bracket assemblies (103), one on forward side with flange aft at hole positions 72 and 73, and one on aft side with flange aft at hole positions 74 and 75. Bracket overlaps existing P&W bracket at hole 75. Place spacing washer (103A) under bracket at hole 74.
 - (p) Install bracket assembly (104) on aft side with flanges aft.
- (9) Installation of brackets on flange "J1": (See view 10.)
- (a) Install bracket assembly (105) on aft side with flange aft. Attach with bolts (105A), washers (105B) and nuts (105C).
 - (b) Rivet nutplate (106D) to bracket (106) using rivets (106E). Install bracket (106) on aft side with flanges aft. Attach bracket with bolts (106A), washers (106B) and nuts (106C).
 - (c) Install bracket assembly (107) on aft side with flanges forward. Attach with bolts (107A), washers (107B) and nuts (107C).
 - (d) Install bracket assembly (108) on aft side with flange aft. Attach with bolts (108A), washers (108B) and nuts (108C).
- NOTE: Make electrical bond between bracket and engine flange. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).
- (e) Install bracket assembly (109) on aft side with flanges aft. Attach with bolts (109A), washers (109B) and nuts (109C).
 - (f) Install bracket (110) on aft side with flanges aft using bolts (110A), washers (110B) and nuts (110C).
- NOTE: Make electrical bond between bracket and engine flange. Maximum allowable resistance across faying surfaces is 0.0025 ohm (2.5 milliohms).
- (g) Install bracket (111) on forward side using bolts (111A), washers (111B) and nuts (111C).

- (10) Installation of brackets on flange K (view 11):
- (a) Install bracket (112) on forward side at top centerline of engine.
 - (b) Install six bracket assemblies (113) on aft side with flanges aft.
 - (c) Install brackets (114A and 114B) as shown in detail M, and brackets (114D and 114E) as shown in detail L. Use bolts (115) and nuts (115A). Install spacer, bolt, washer, nut, and cotter pin to complete bracket assemblies (114 and 114C).
 - (d) Install bracket (116) on aft side.
 - (e) Install bracket (117) on forward side with flanges forward. Install five detector bonding bracket (117A) on aft side using existing bolts tightened to 110-150 pound-inches. Maximum allowed resistance between faying surfaces is 5 milliohms.
 - (f) Install bracket assembly (118) on aft side with flange aft and bracket assembly (119) on forward side. Two bolts are common to both.
 - (g) Install bracket assembly (120) on forward side with flange forward, and install bracket assembly (121) on aft side with flange aft. Two bolts are common to both.
 - (h) Install five bracket assemblies (122) on forward side with flanges forward.
 - (i) If used, install bracket assembly (123) on forward side with 115-degree angle flange forward. Otherwise, install bracket assembly (123A) on aft side with flange aft.
 - (j) If used, install bracket assembly (124) on aft side with shorter flange forward. Otherwise, install bracket assembly (124A) on aft side with flange aft.
- (11) Installation of brackets on rear engine mount flange (view 12):
- (a) Install bracket (125) between rear engine mount flanges in two places as shown in detail N. Use washers (129) between bracket and flange as necessary to prevent preloading engine flange. Secure with bushings (126), bolts (127), washers (130) and nuts (128).
 - (b) Install rub strip (131) on rear surface of flange as shown in detail P. Place washer (133) between rub strip and engine flange. Attach with bolt (132), washers (134), and nut (135).

(c) If required, install bracket (136), bushing (137), and bushing (138) as shown in detail Q. Attach with washers (139), bolts (140), and nuts (141).

- (12) Locate bracket assembly (142) at lower centerline of engine inlet accessory flange. Remove two Pratt and Whitney flange bolts and reinstall to attach bracket (view 13).
- (13) Locate bracket assembly (143) on TAI valve at top centerline of engine aft of flange C. Remove two TAI flange bolts and reinstall to attach bracket (view 14).
- (14) Install bracket (144), bracket assembly (145), bracket (146), and bracket assembly (147), with screws (148), washers (149), and nuts (150). Locate parts as shown in view 15.

NOTE: Make electrical bond between all parts. Maximum allowable resistance between any part and basic structure in 0.0025 ohms (2.5 milliohms).

- (15) Install bracket (151) on forward face of fuel control mount flange. Remove two flange bolts and replace to attach bracket (view 16).
- (16) Install quick-release clamp assembly (152) on bracket (151) with two screws (153) and nuts (154).
- (17) Install bracket assembly (155) on right bleed port cover as shown in view 17. Remove two Pratt and Whitney cover attach bolts and reinstall to attach bracket. Lockwire bolts in place.
- (18) Install current transformer support on right side between flanges G and H as follows (view 18):

NOTE: Make electrical bond between all brackets and basic structure. Maximum allowable resistance between any bracket and basic structure is 0.0025 ohms (2.5 milliohms).

- (a) Attach bracket (156) to aft side of bracket (67) on flange G with two screws (157) and nuts (158).
- (b) Attach bracket (159) to forward side of bracket (74) on flange H with two screws (157) and nuts (158).
- (c) Attach bracket assembly (160) to brackets (156 and 159) with screws (161), nuts (162) and washers (162A).

(19) Install bracket (163) on left bleed pad cover as shown in view 19. Remove two Pratt and Whitney cover attach bolts and reinstall to attach bracket. Lockwire bolts in place.

(20) Install brackets in gearbox area as follows (view 20):

(a) Install bracket (164) as shown in detail R. Install spacer (166) between bracket and gearbox surface, and attach bracket with nuts (165). Attach bracket assy (166A) on bracket assy (164) with screws (166B) and washers (166C).

(b) Install bracket (167) on aft gearbox flange with three existing Pratt and Whitney flange bolts. Install one or two washers (168) between bracket and gearbox flange at each bolt.

NOTE: Make electrical bond between bracket and gearbox. Maximum allowable resistance across faying surfaces is 0.0025 ohms (2.5 milliohms).

(c) Install nutplate (169) or bracket (169A) on bracket (167).

1) 65-45798-6 -- install bracket (171A) using bolt (171B), washer (171C) and nut (171D).

2) All other installations -- install nutplate (169) using screws (170), nuts (171) and washers (168). Make electrical bond as in step (b).

(d) Install bracket (172 or 174A) as shown in detail T. Attach with bolts (173 or 174B) and washers (174 or 174C). Make electrical bond as in step (b). Lockwire bolts in place.

(e) Install angle (175) with bolt (176) and washer (177). Make electrical bond as in step (b).

(f) Install bracket (178) on front flange of gearbox as shown in detail U. Remove two Pratt and Whitney flange bolts and reinstall to attach bracket.

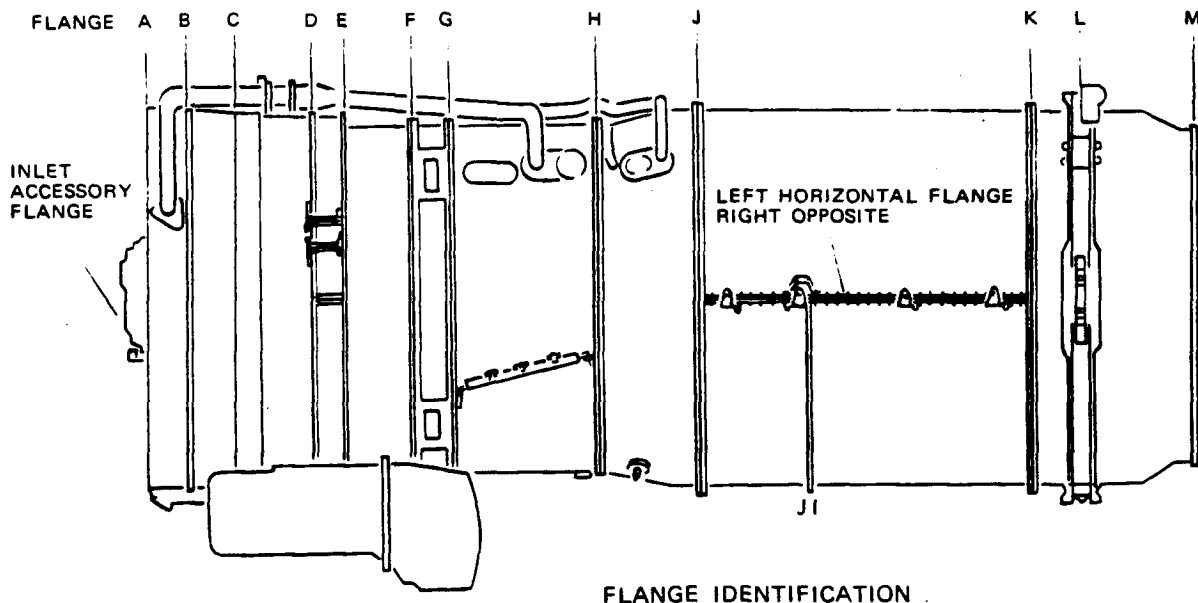
(g) Install bracket assembly (179) on forward flange of gearbox as shown in detail U. Remove two Pratt and Whitney flange bolts and reinstall to attach bracket assembly.

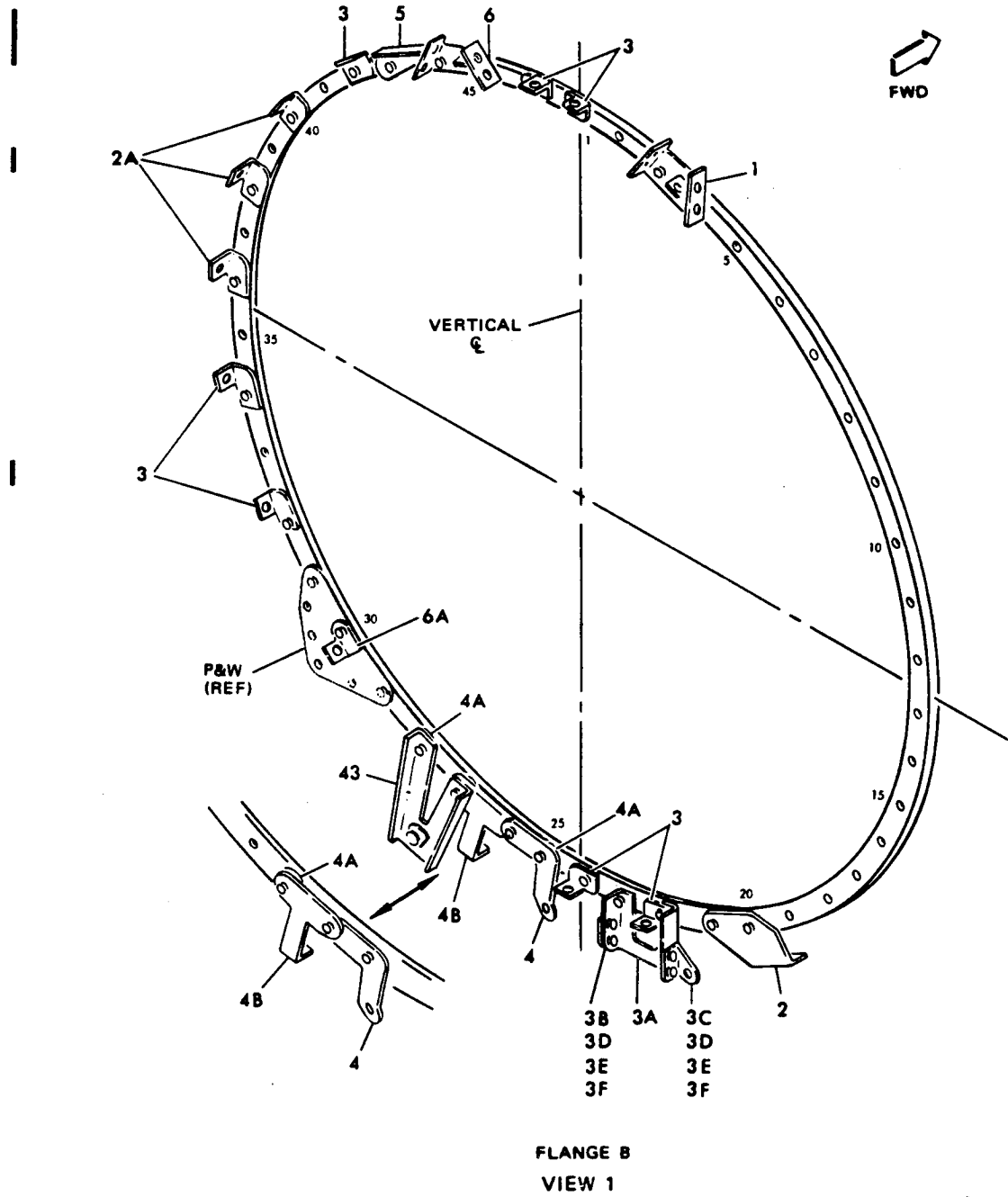
(h) Install quick-release clamp assembly (180) on bracket assembly (179) with screws (181) and nuts (182) (detail U).

(i) Install bracket (183) on forward flange of gearbox. Remove two Pratt and Whitney flange bolts and reinstall to attach bracket (detail U).

- (21) Locate bracket assemblies (184 and 185) as shown in view 21.
Attach with existing Pratt and Whitney bolts.
- (22) Install CSD oil filter support brackets as follows: (See view 22.)
 - (a) Locate bracket (186). Attach to bracket (78) on flange "H" and bracket (97) on flange "J" with screws (187), washers (188), and nuts (189).
 - (b) Attach bracket (190) to aft, inside leg of bracket (186) with screw (191), washer (192), and nut (193).
 - (c) Attach bracket (194) to forward, inside leg of bracket (186) with screw (195).
 - (d) Attach bracket assembly (196A) to bracket (186) using screw (196B), washer (196C), and nut (196D).
- (23) Install brackets on left horizontal flange as follows: (See view 23.)
 - (a) Install bracket assembly (197) in three places on bottom face of flange. Attach with existing Pratt and Whitney flange bolts.
 - (b) Install bracket assembly (198) on bottom face of flange. Attach with existing Pratt and Whitney flange bolts.
 - (c) Install two bracket assemblies (199) on bottom face of flange. Attach with existing Pratt and Whitney flange bolts.
 - (d) Install bracket assembly (200) on top of ignition box support bracket with two screws (201), nuts (202) and washers (202A).

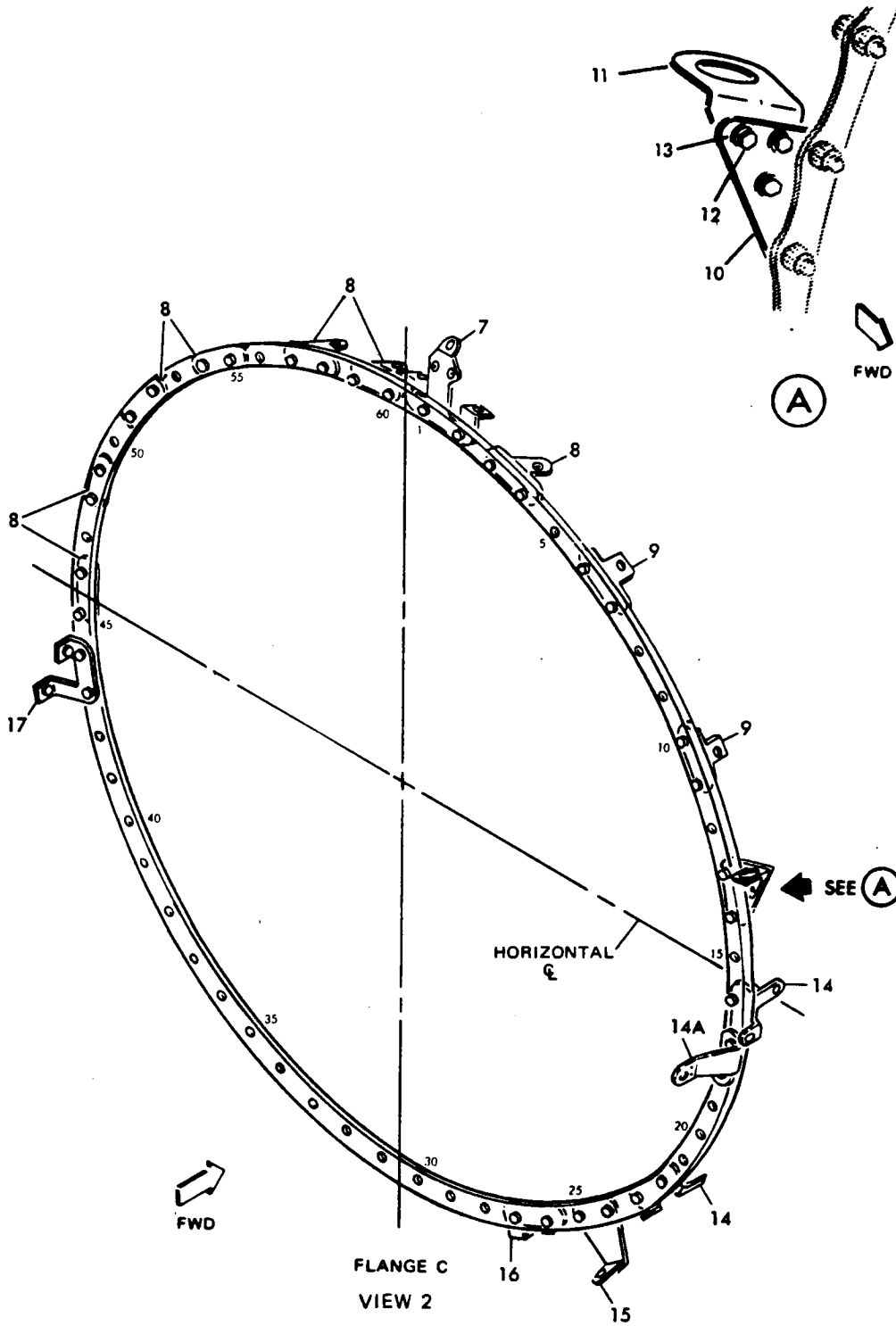
- (24) Install brackets on right horizontal flange as follows (view 24):
- (a) Install bracket assembly (203) on bottom face of flange. Attach with four existing flange bolts.
 - (b) Install bracket (204) on upper face of flange. Attach with existing flange bolts.
 - (c) Install bracket (205) on bottom face of flange. Attach with existing flange bolts.
 - (d) Install bracket assembly (206) on bottom face of flange. Attach with existing flange bolts.
 - (e) Install brackets (207 and 208) on upper face of flange. Attach with existing flange bolts.
 - (f) Locate bracket (209) and attach to brackets (207, 208) with screws (210), washers (211), and nuts (212). Attach to forward side of flange J1 with bolts (213), washers (214), and nuts (215).
- (25) Locate support (216, 216A) as shown in view 25. Attach to brackets (91, 106) with screws (217, 217A), nuts (218, 218A) and spacers (218B).
- (26) Locate fire detector support (219) between flanges J1 and K as shown in view 26. Attach to bracket (108) at flange J1 and to drain support bracket mounted on bracket (117) at flange K with four screws (220), washers (221), and nuts (222).



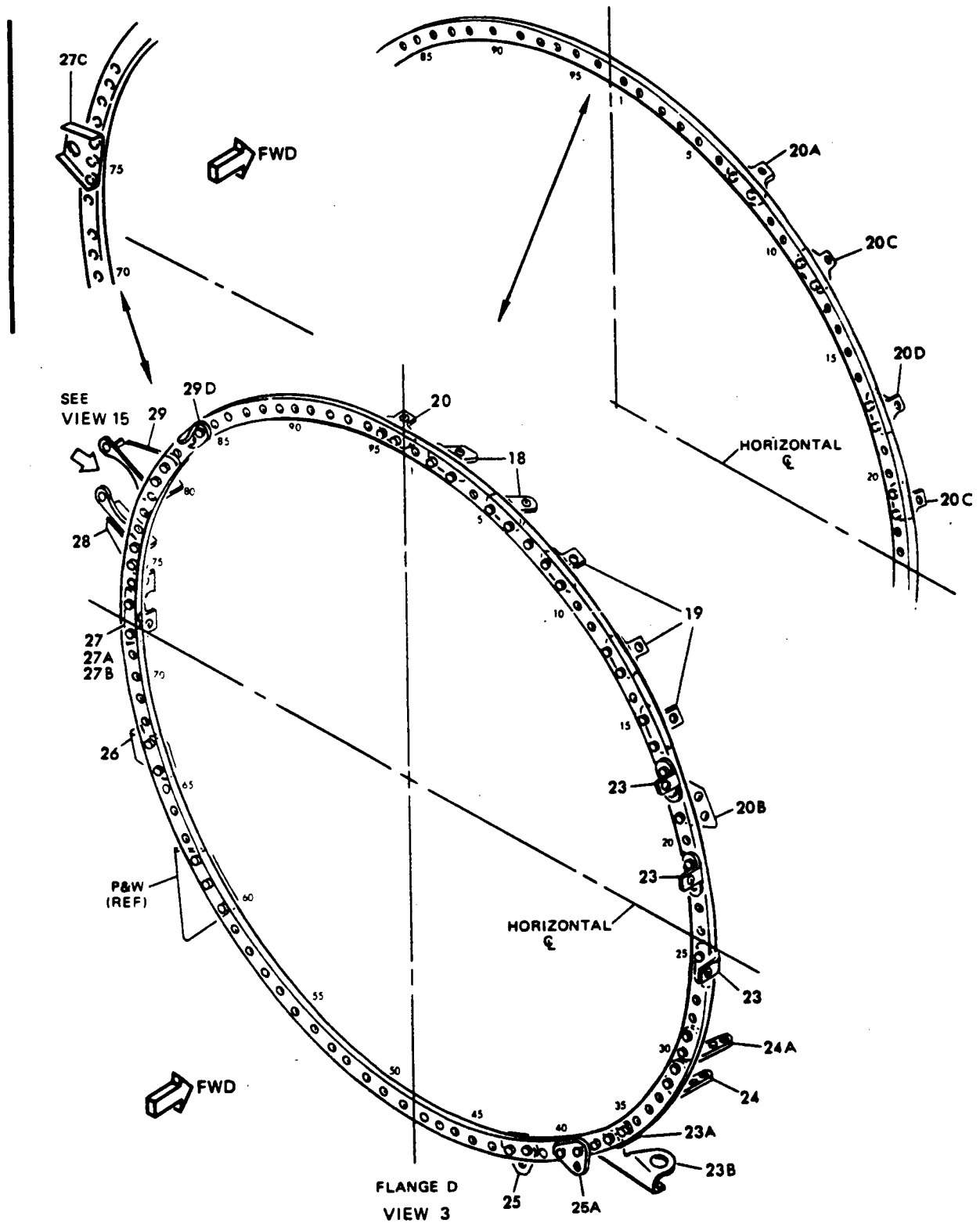


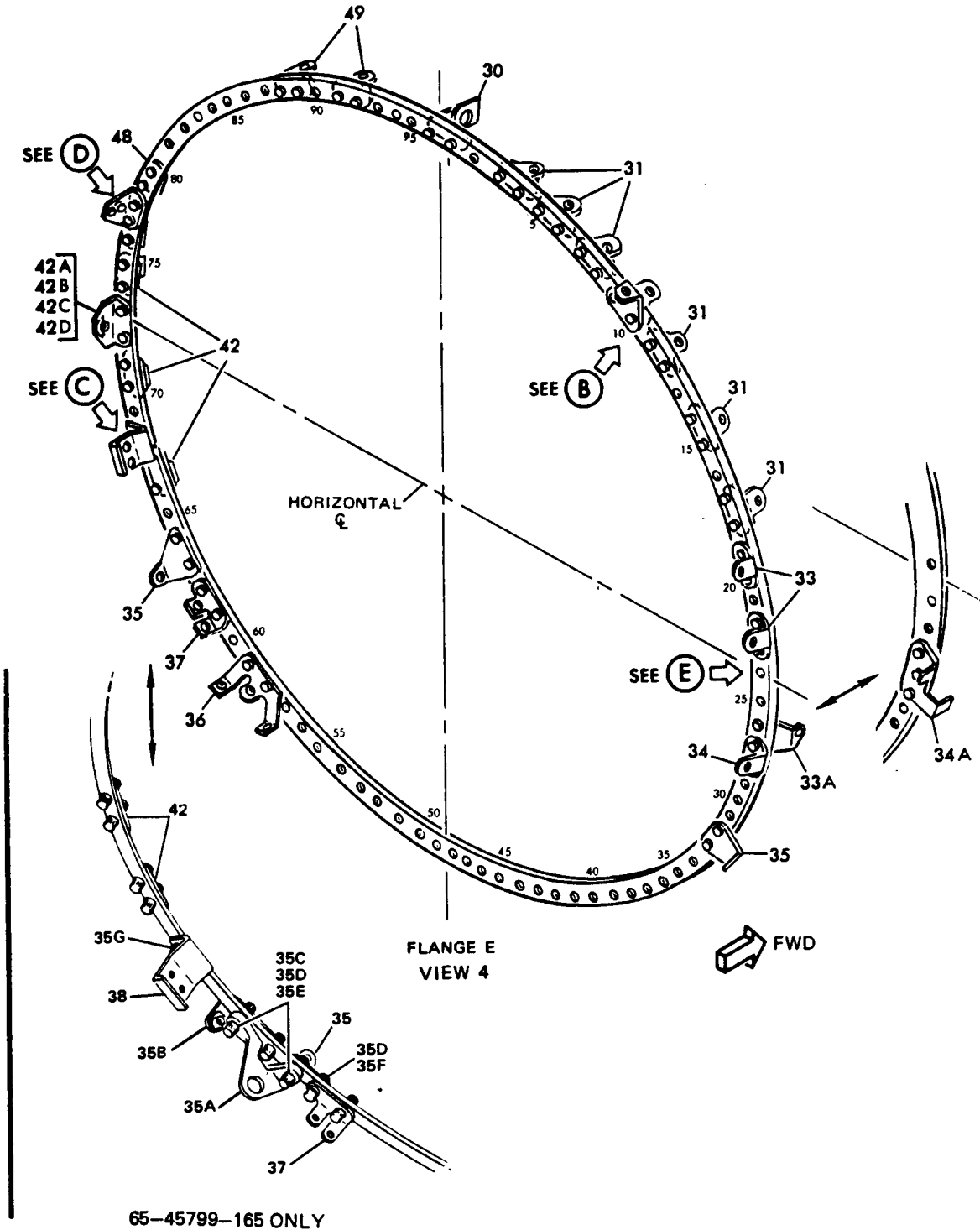
Jun 10/71

Engine Accessory Support Bracket Installation
Figure 520 (Sheet 1)

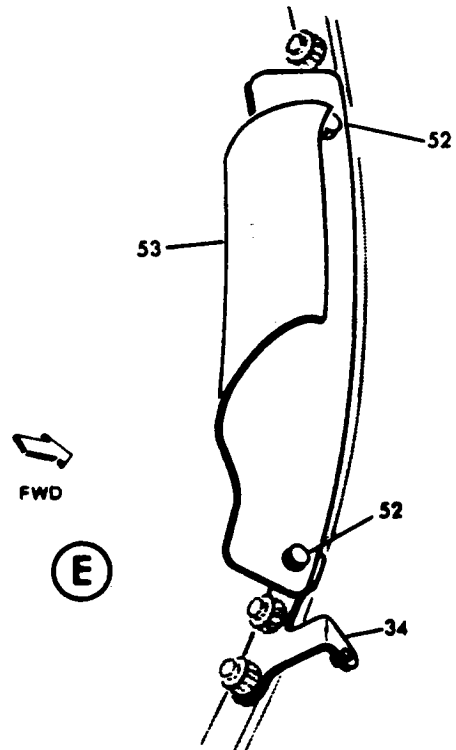
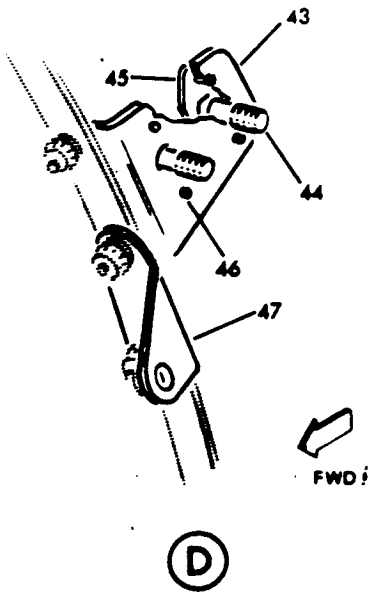
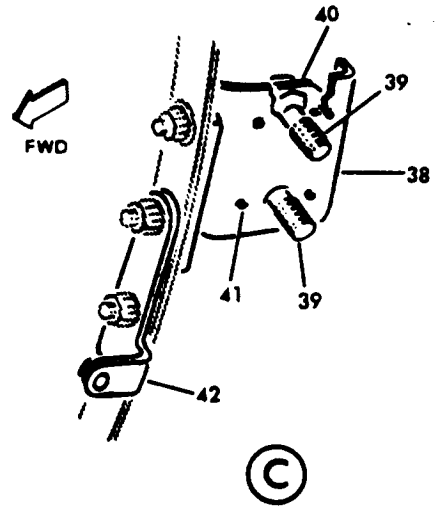
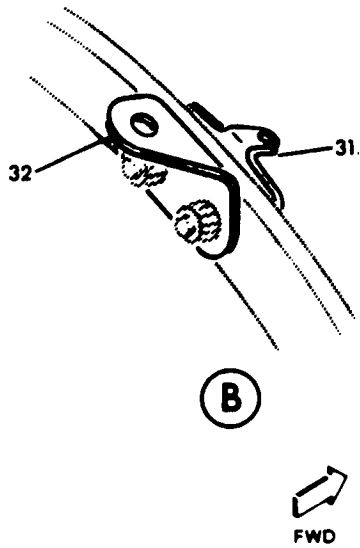


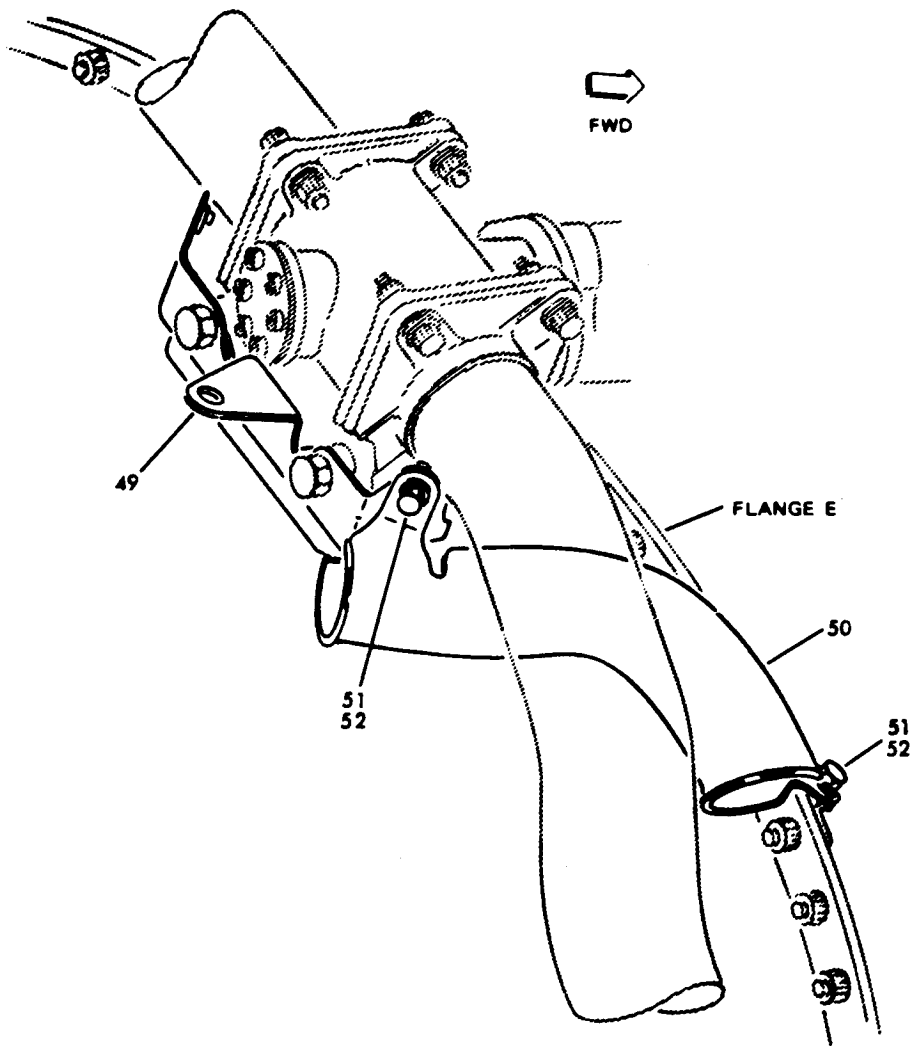
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 2)





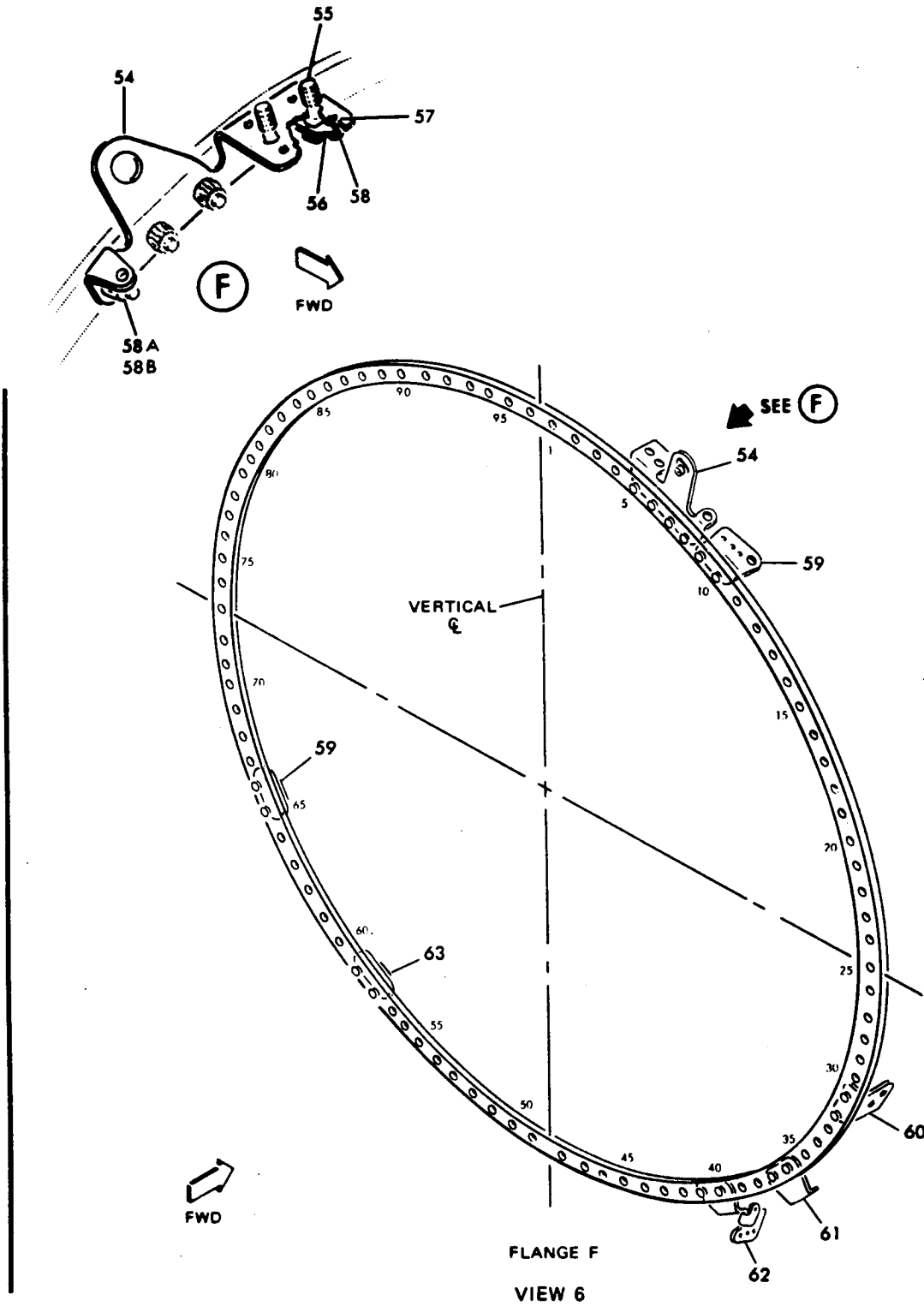
Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 4)

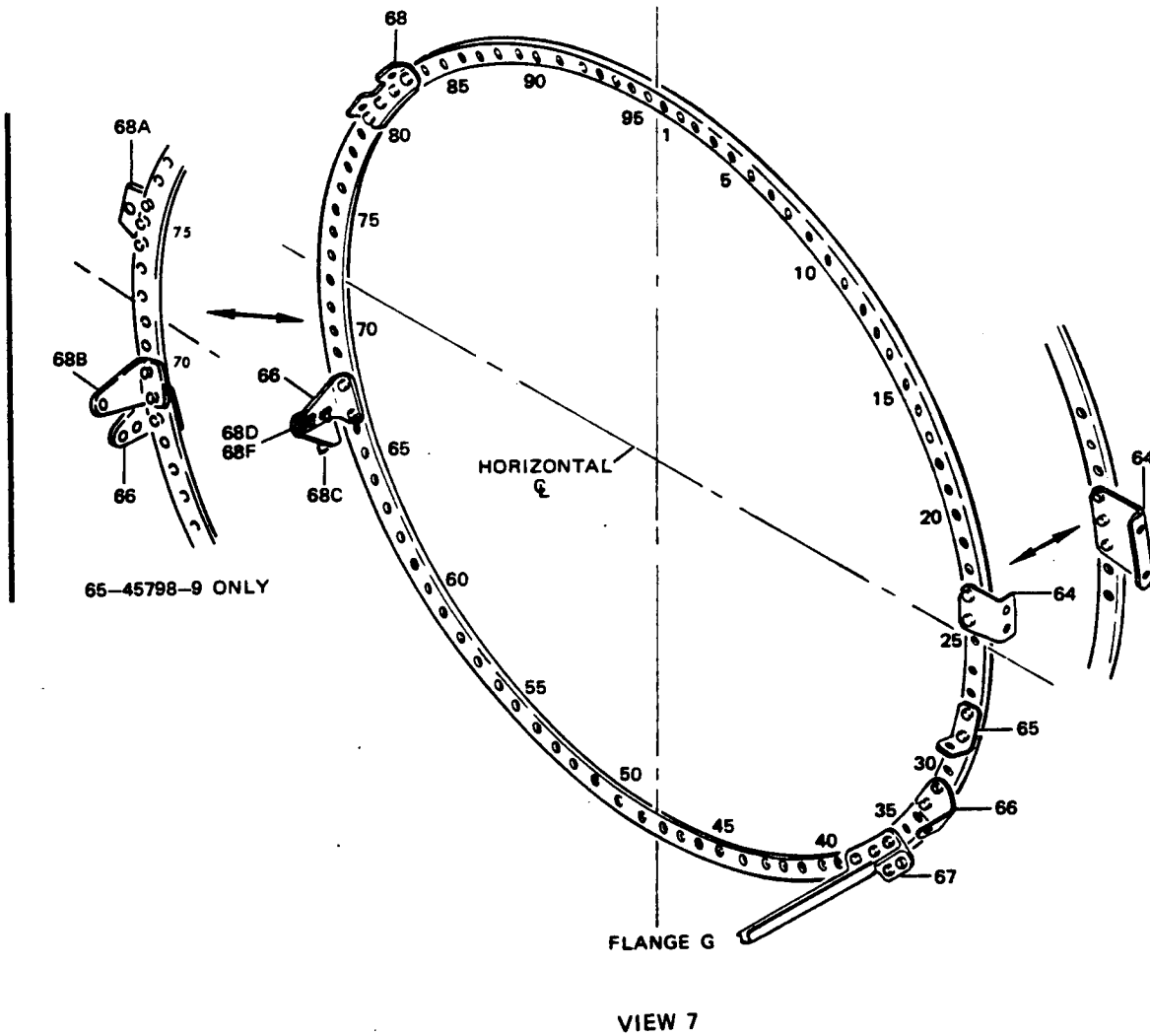




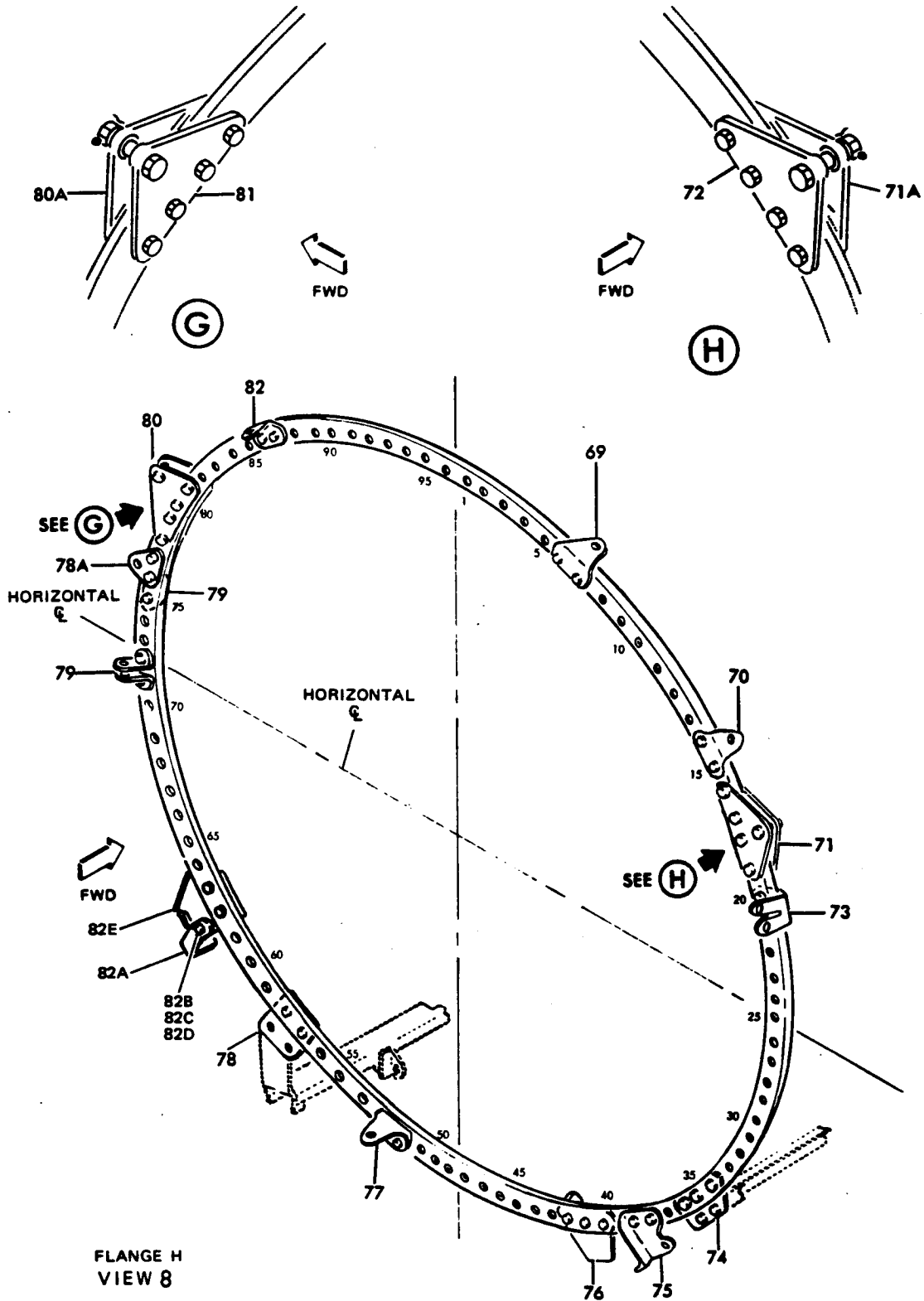
RIGHT SIDE

VIEW 5

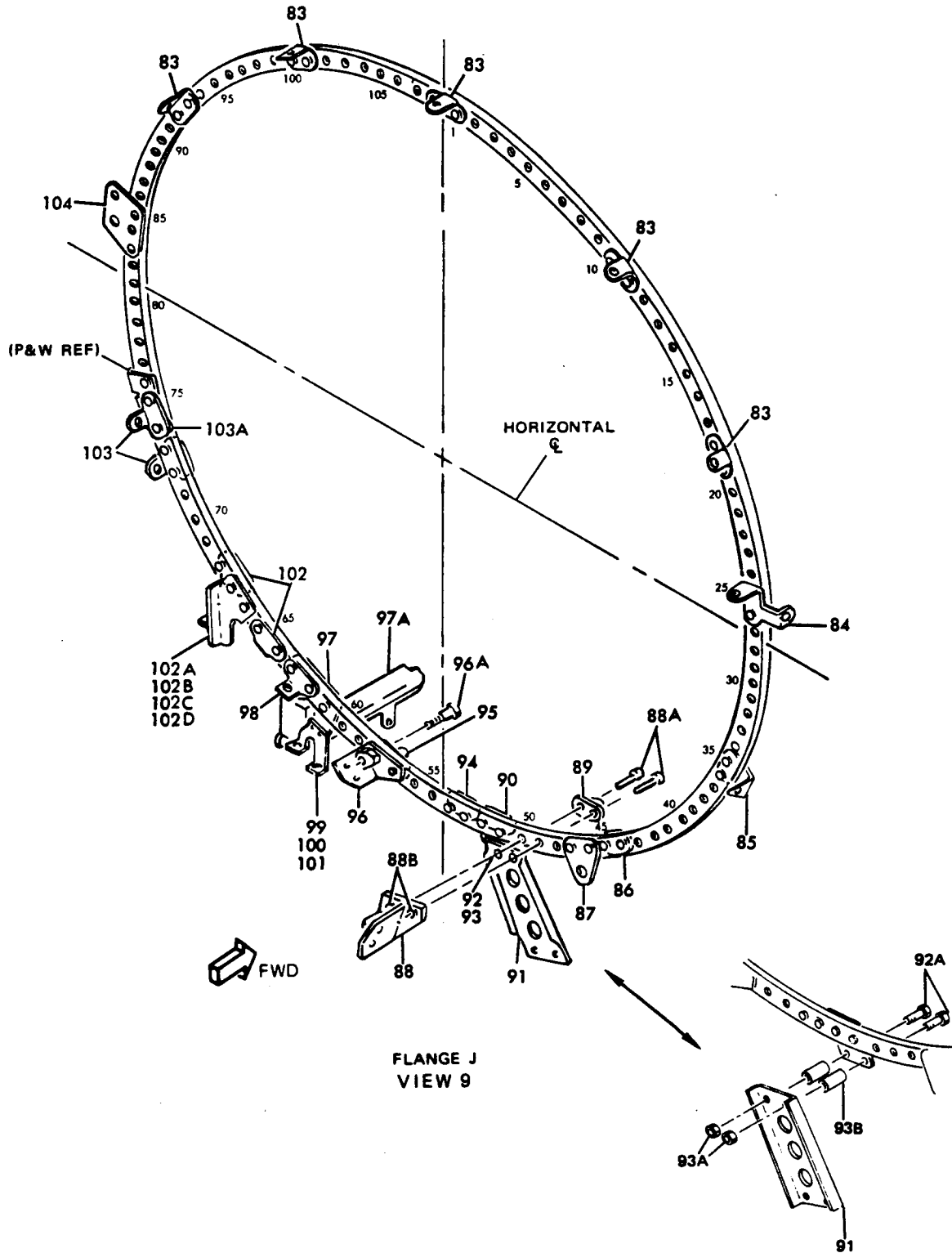




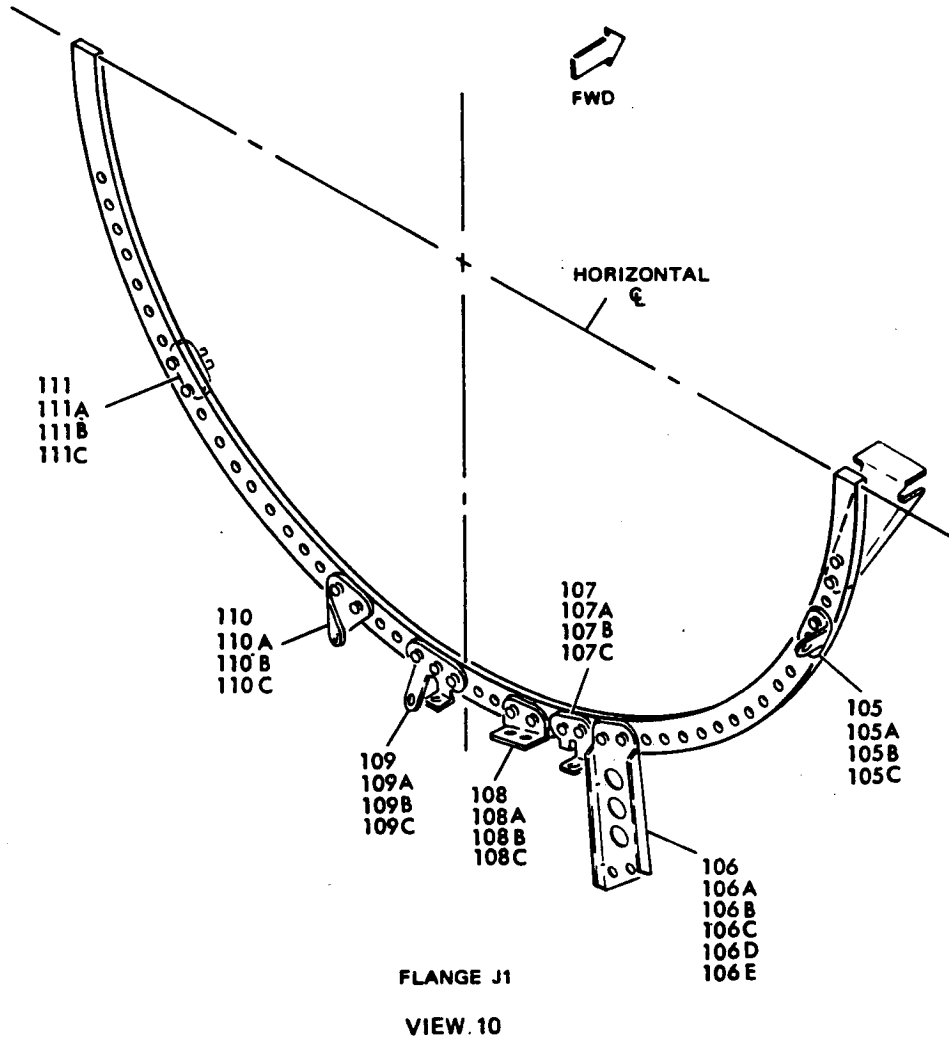
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 8)



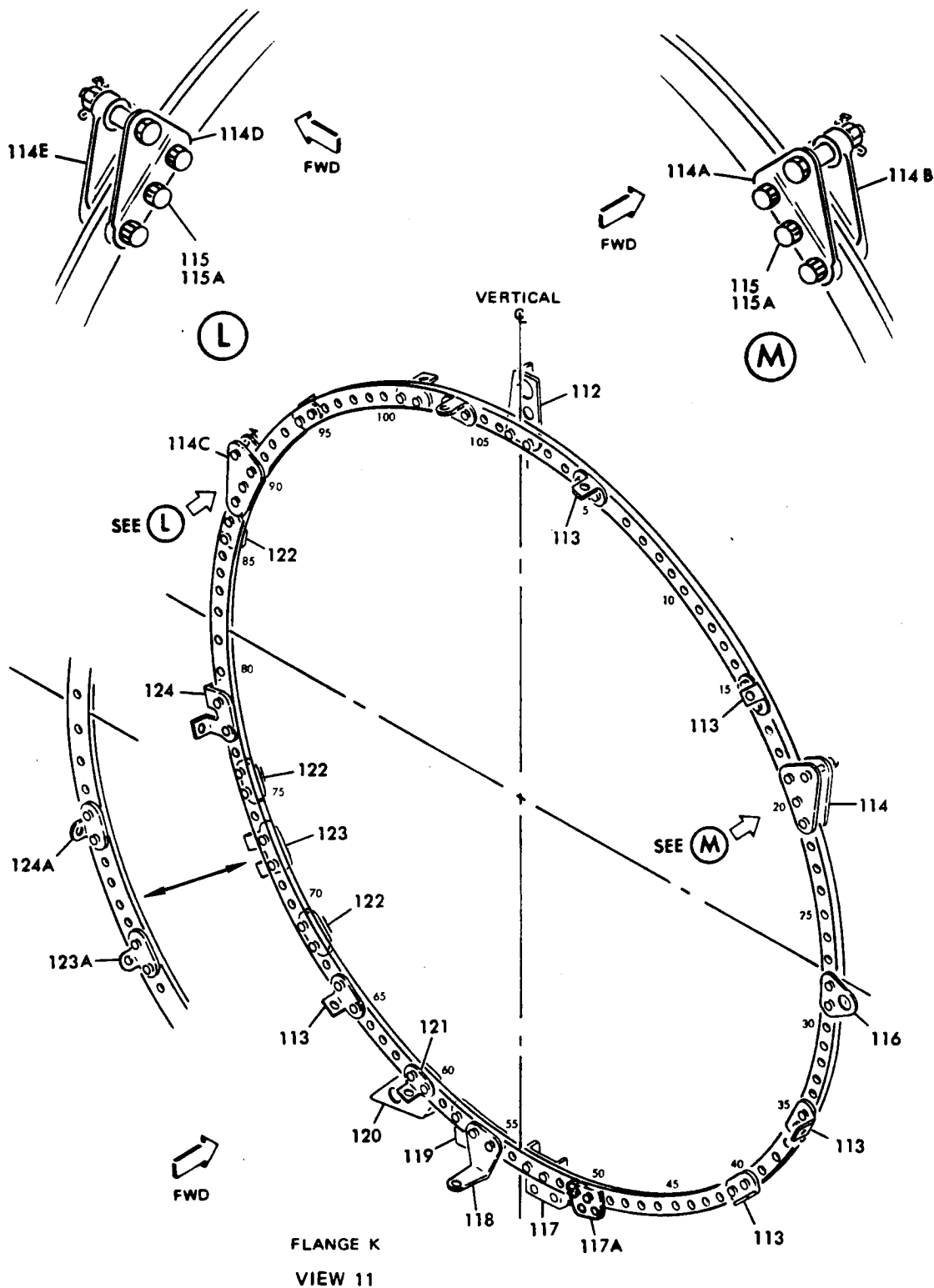
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 9)



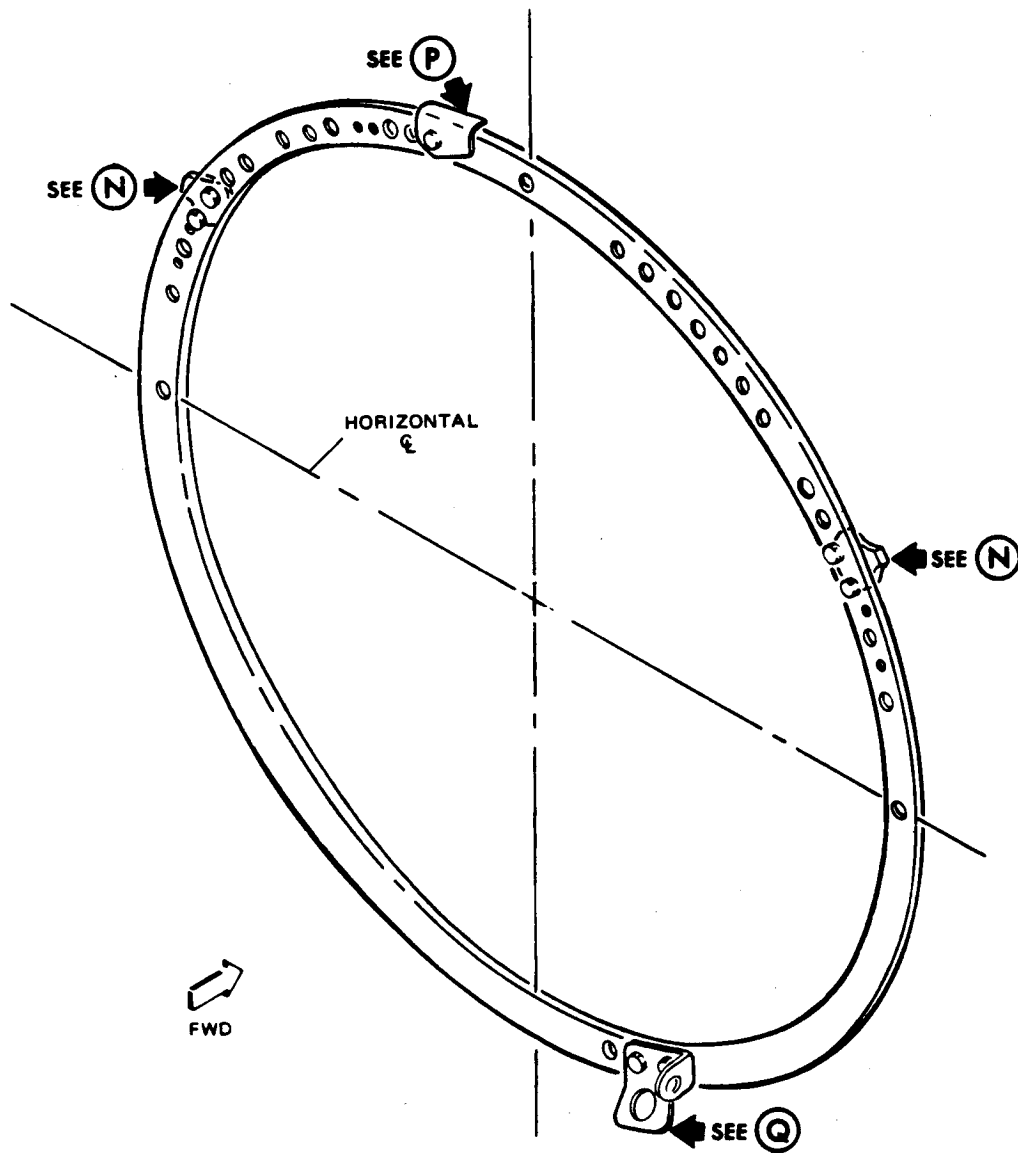
Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 10)



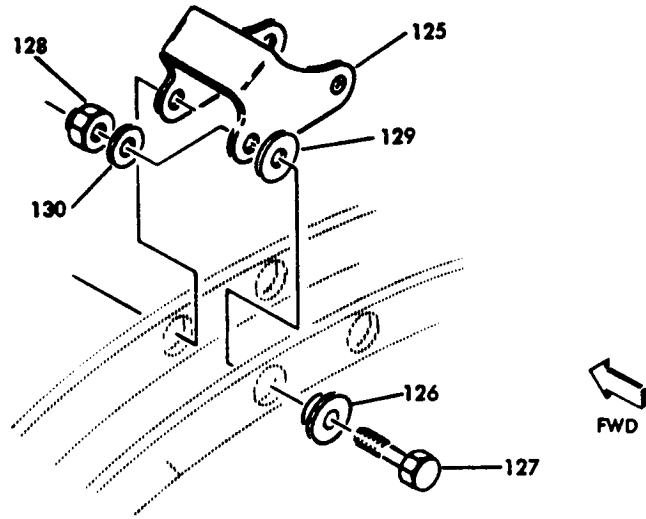
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 11)



Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 12)

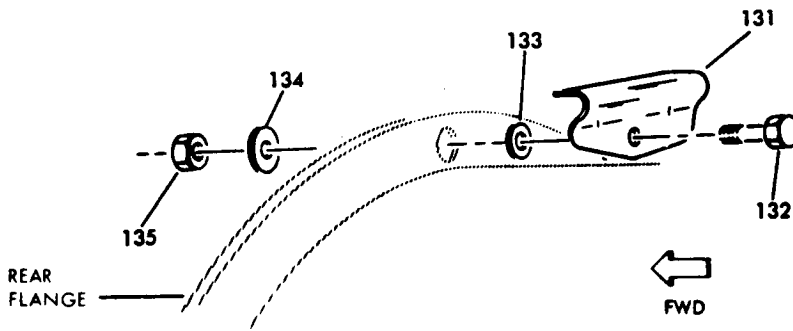


VIEW 12
REAR ENGINE MOUNT FLANGES

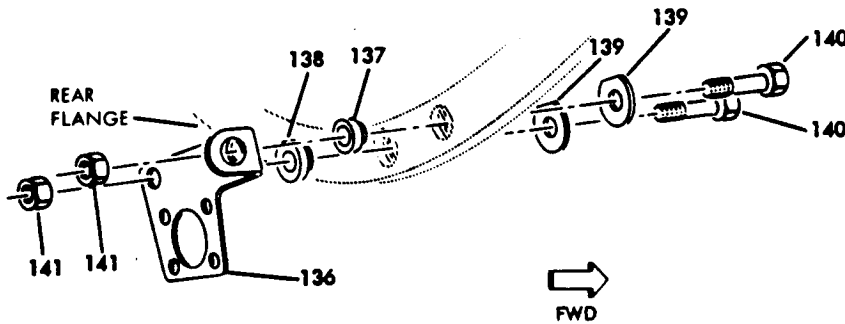


LEFT SIDE SHOWN
 RIGHT SIDE OPPOSITE

(N)



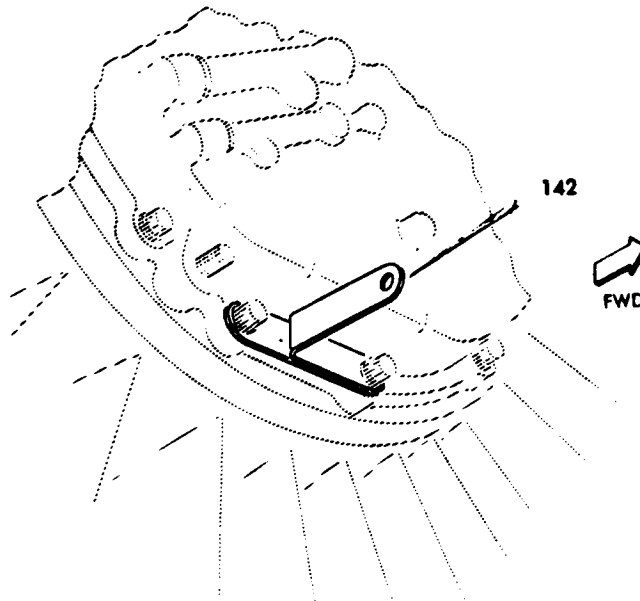
(P)



(Q)

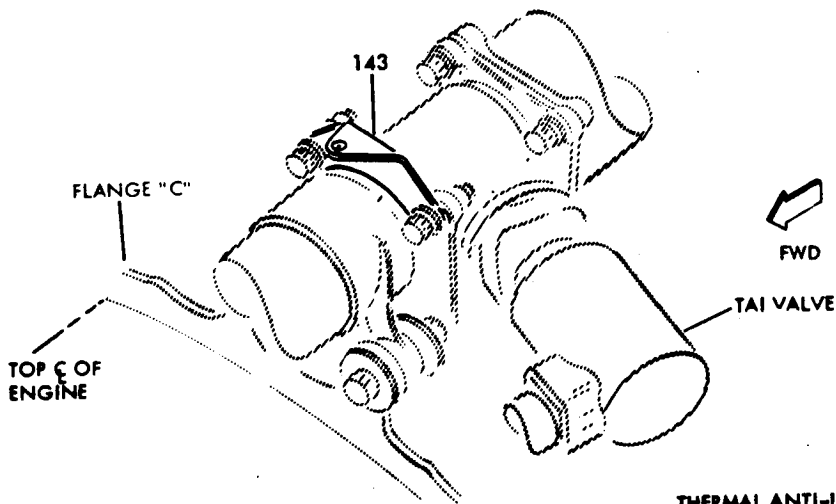
Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 14)

May 15/68



INLET ACCESSORY FLANGE

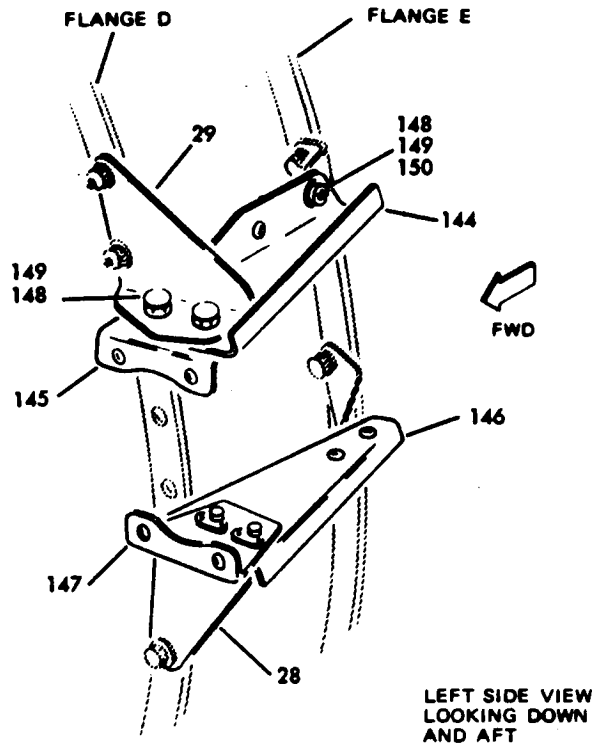
VIEW 13



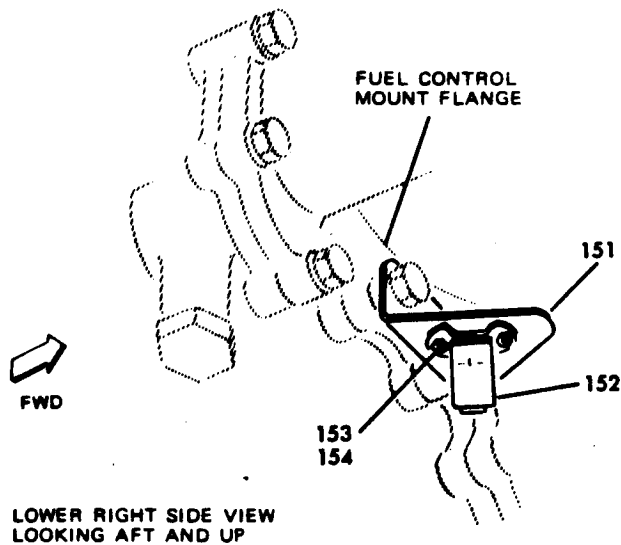
THERMAL ANTI-ICING
VALVE VIEW LOOKING
DOWN AND AFT

VIEW 14

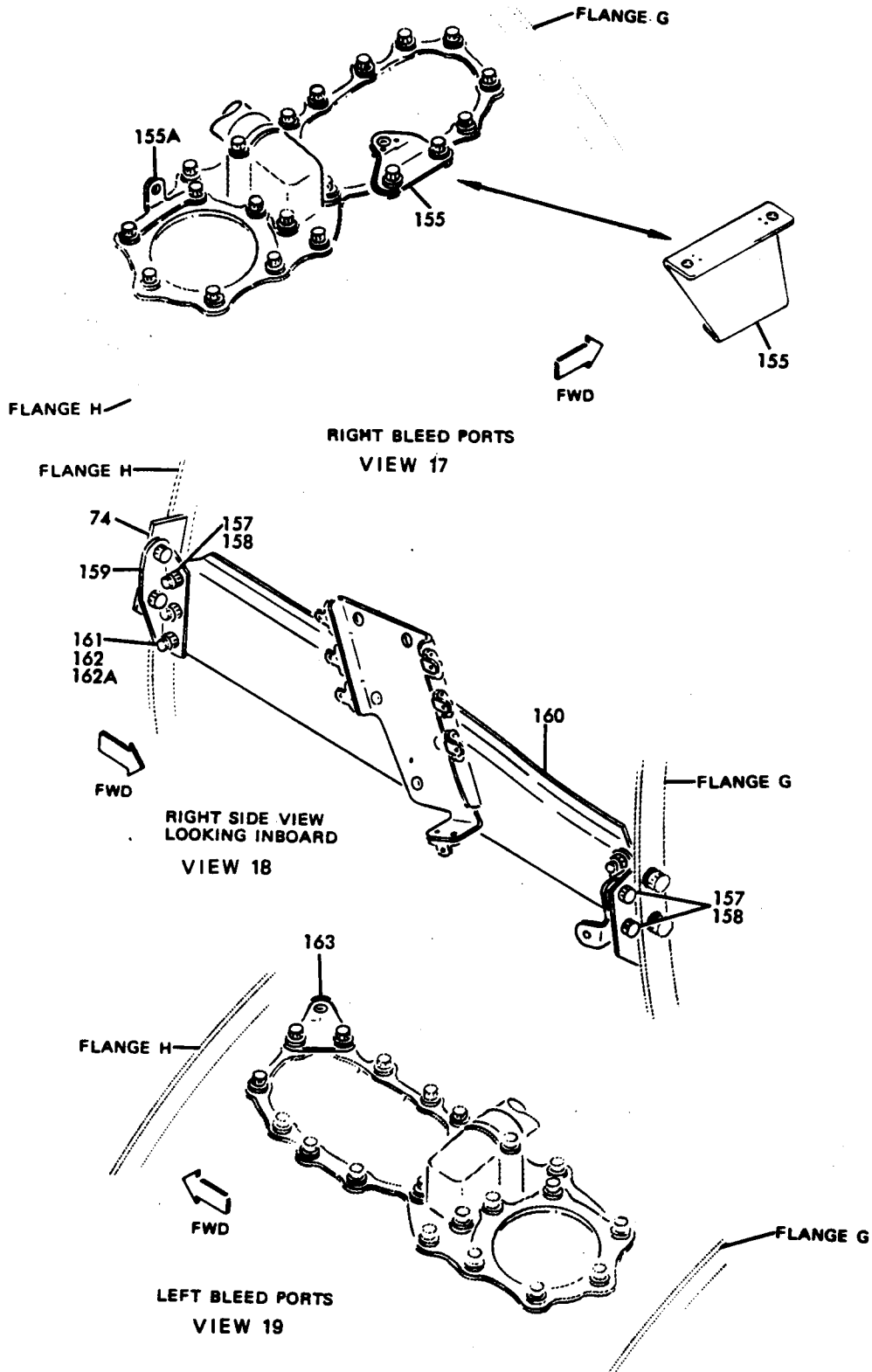
May 15/68



VIEW 15

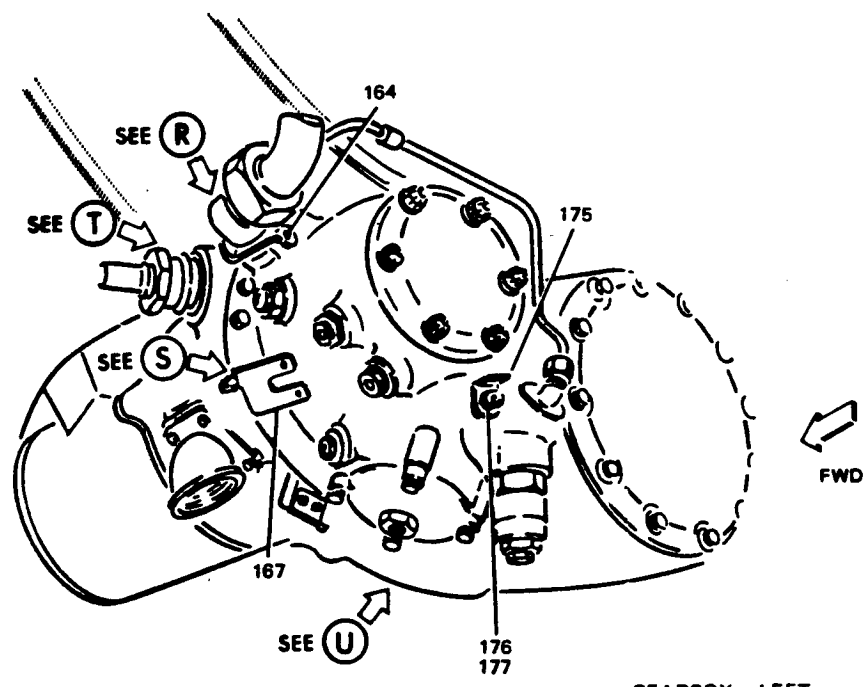
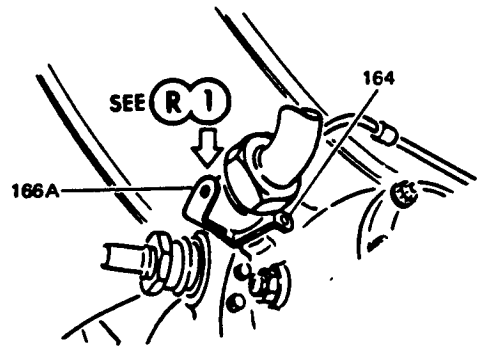


VIEW 16



Nov 10/69

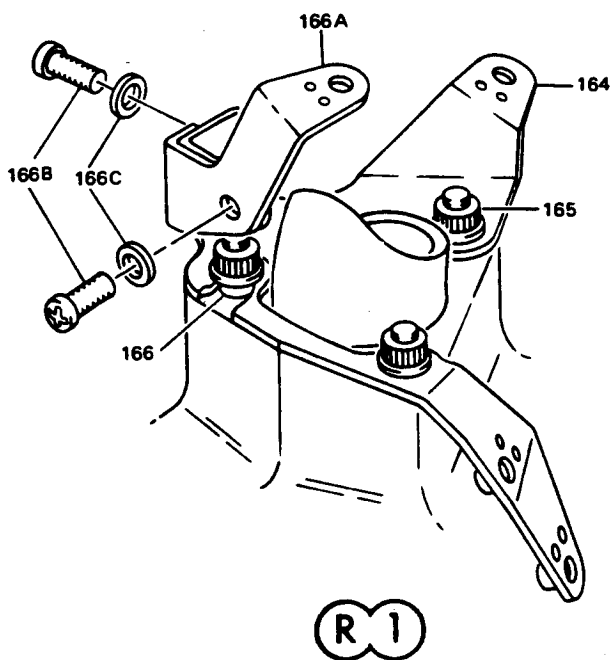
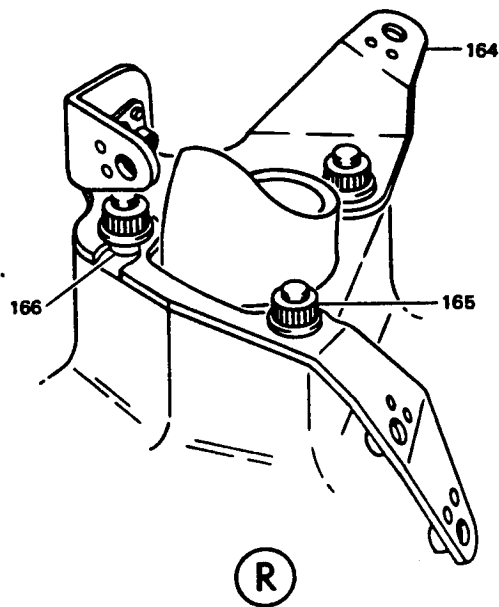
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 17)



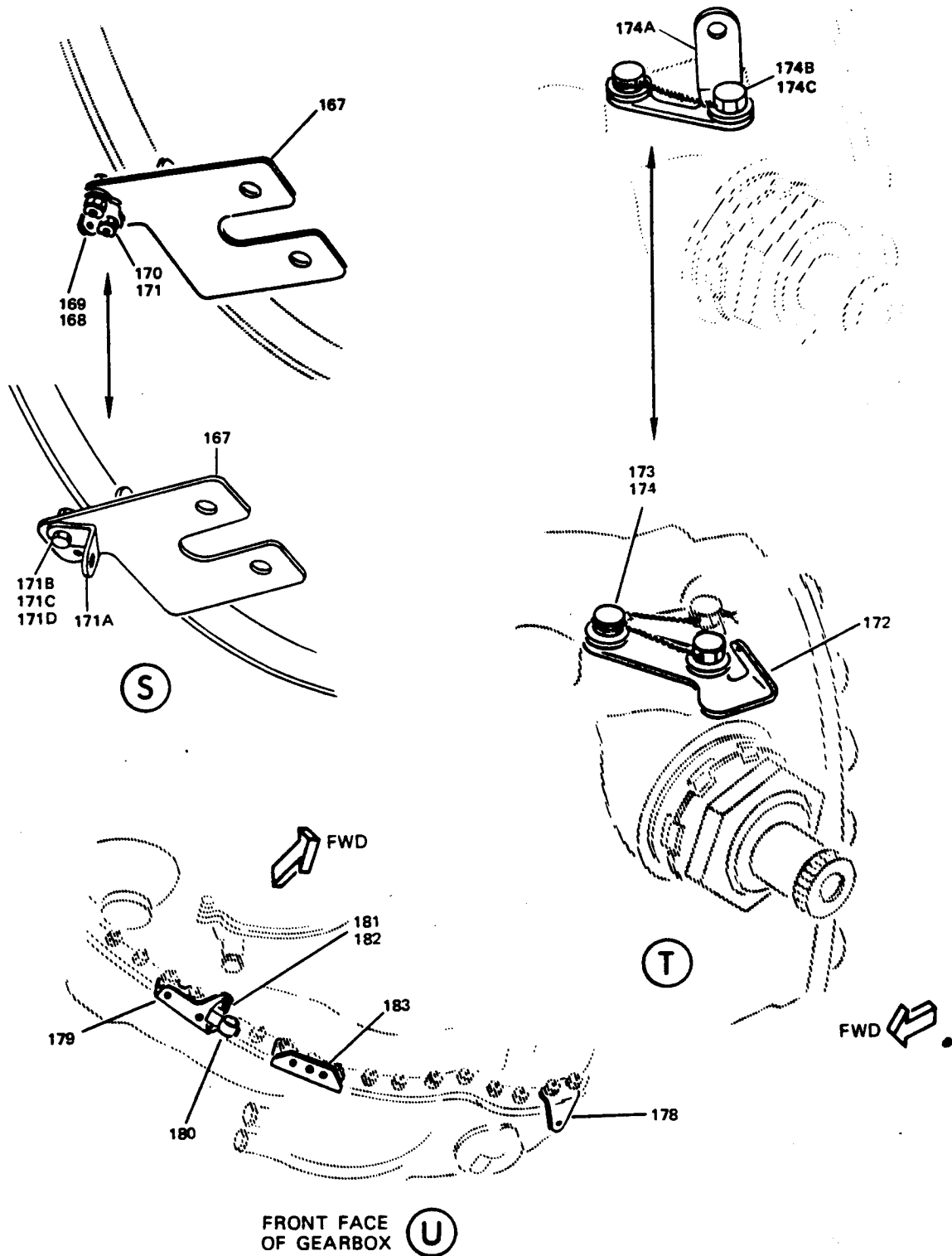
GEARBOX - LEFT
SIDE VIEW LOOKING
UP AND FORWARD

VIEW 20

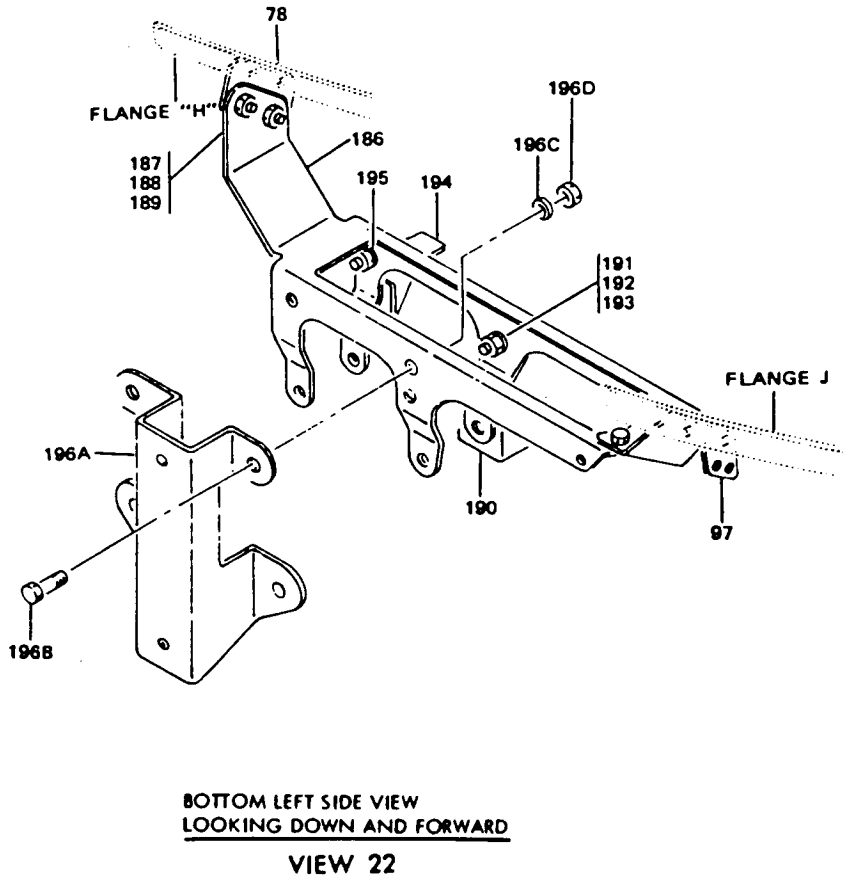
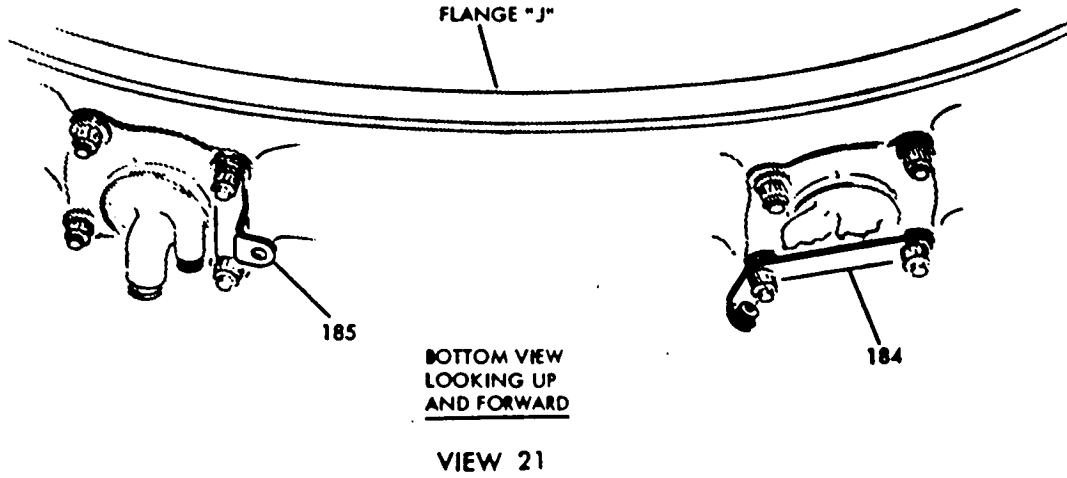
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 18)



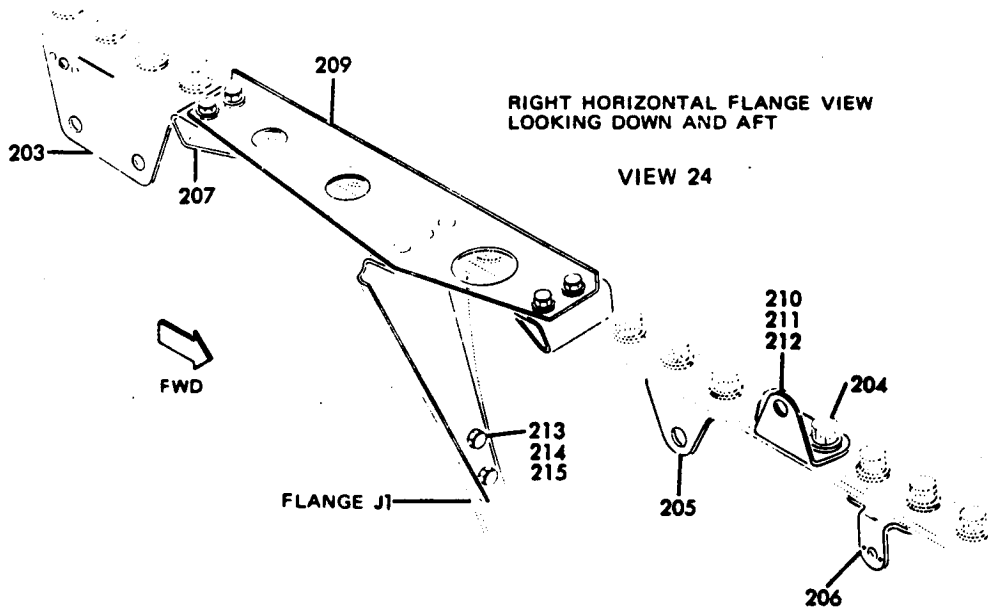
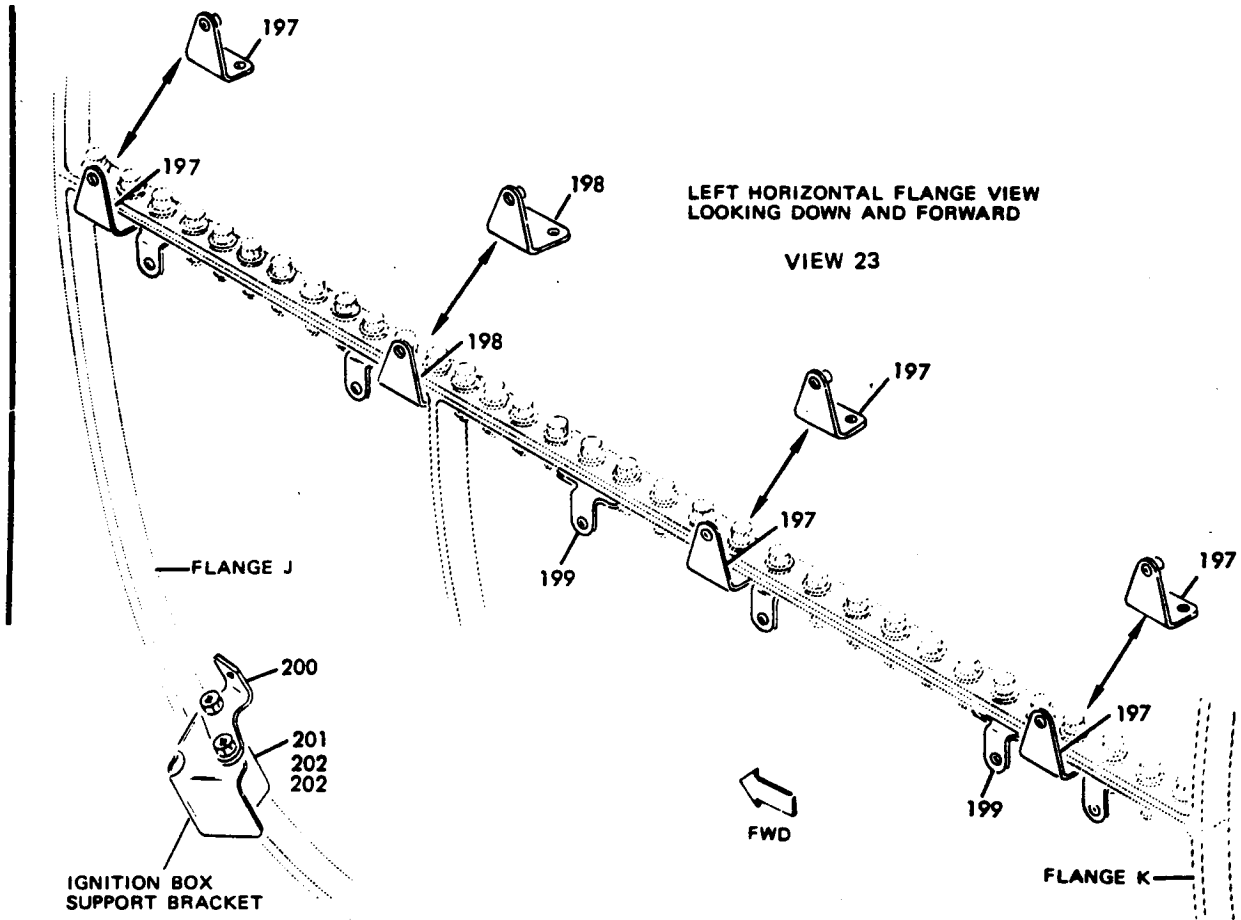
Engine Accessory Support Bracket Installation
Figure 520 (Sheet 18A)

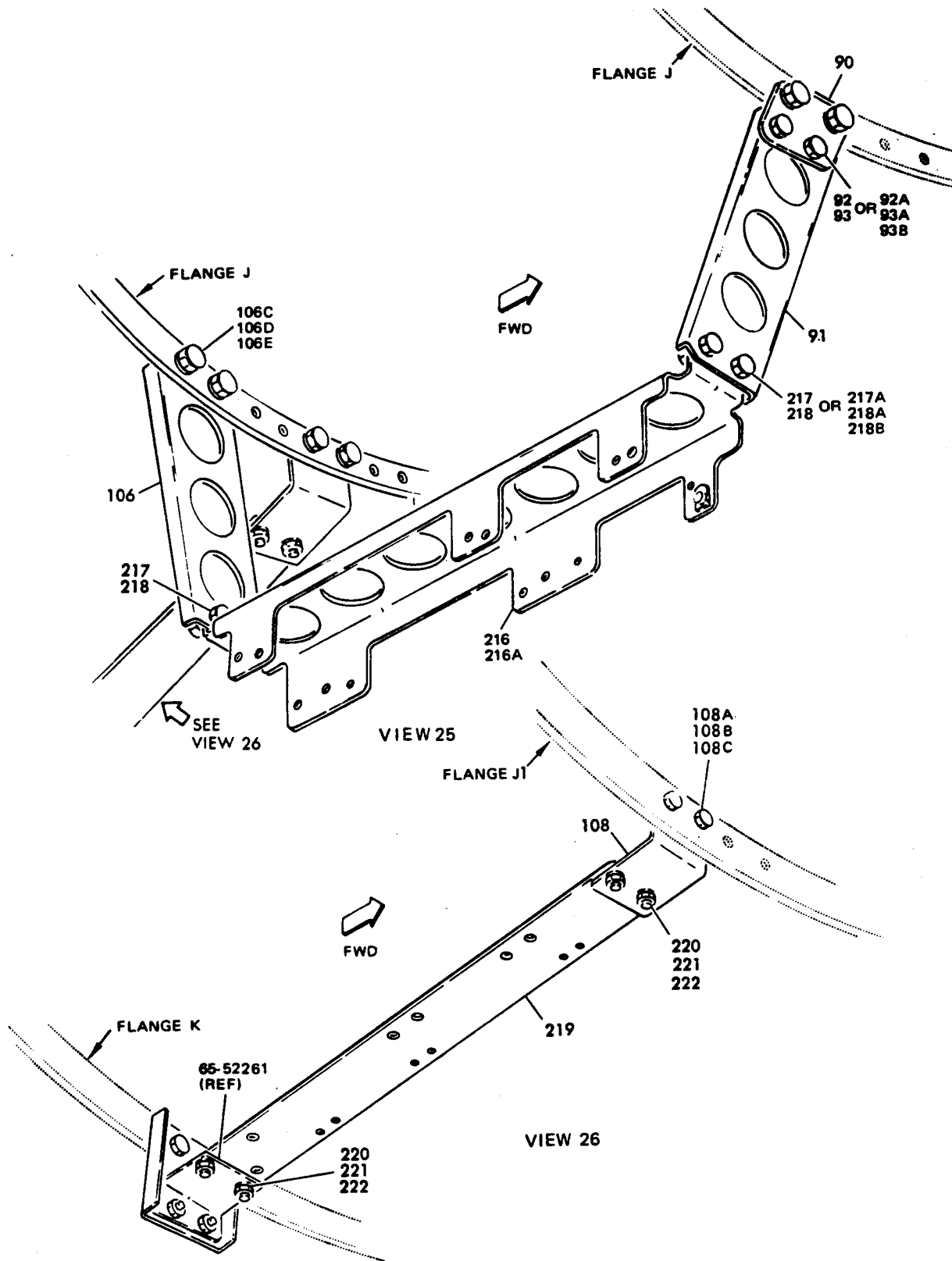


Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 19)



Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 20)





Engine Accessory Support Bracket Installation
Figure 520 (Sheet 22)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY		
A.	(1)(a)	1	BRACKET	464459		Ref	
		2	BRACKET	65-45799-81		1	
	(b)	2A	BRACKET ASSY	65-22486-30 (used on 65-45799-1, -117, -120, -127)		3	
		2A	BRACKET ASSY	463974 (used on 65-45799-1, -117, -120, -127)		3	
		3	BRACKET ASSY	65-22486-30		7	
	(c)	3A	BRACKET ASSY	463974		7	
			BRACKET	65-22486-121 (used on 65-45799-117, -120, and -127)		1	
	(d)	3E	BRACKET	69-60758-1 (used on 65-45798-1 only)		1	
			BRACKET	69-60758-2 (used on 65-45798-2 only)		1	
		3D	BOLT	BACB3ONE3-3		4	
		3E	NUT	BACN1OJC3		4	
	(e,f)	3F	WASHER	AN960-10		4	
			4	BRACKET	65-45799-29		1
			4A	WASHER	AN960D416		2
4E			BRACKET ASSY	544260		Ref	
4C			BRACKET	65-45799-126 (used on 65-45799-127; optional on 65-45799-117 and -120)		1	
(g)	5	BRACKET	458075		Ref		
		6	BRACKET	491633		Ref	
(h)	6A	BRACKET	65-22486-32		1		
		BRACKET	463983		1		
(2)(a)	7	BRACKET ASSY	65-45799-140		1		
		BRACKET ASSY	65-45799-23 (opt1)		1		
(b)	8	BRACKET ASSY	65-22486-195		7		
		BRACKET ASSY	491641		7		

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(c)	9	BRACKET ASSY	65-45799-43		2
(d)	10	BRACKET	65-45799-805		1
(e)	11	BRACKET ASSY	69-48205-3		1
	12	SCREW	BACS12CB3-9		3
	13	WASHER	AN960-10L		3
(f)	14	BRACKET ASSY	65-45799-41		2
	14A	BRACKET ASSY	65-45799-128 (used on 65-45799-127)		
(g)	15	BRACKET ASSY	65-45799-103		1
	15	BRACKET ASSY	65-45799-59 (opt1)		1
(h)	16	BRACKET ASSY	65-45799-48		1
(i)	17	BRACKET ASSY	65-45799-16		1
(3)(a)	18	BRACKET ASSY	65-22486-137		2
	18	BRACKET ASSY	480389		2
(b)	19	BRACKET ASSY	65-22486-206		3
	19	BRACKET ASSY	491585		3
(c)	20	BRACKET ASSY	65-22486-206		1
	20	BRACKET ASSY	491585		1
(d)	20A	BRACKET ASSY	65-45799-206 (used on 65-45799-117 only)		1
	20A	BRACKET ASSY	491585 (used on 65-45799-117 only)		1
(e)	20B	BRACKET ASSY	65-45799-51 (used on 65-45799-117 only)		1
(f)	20C	BRACKET ASSY	65-45799-65 (used on 65-45799-123 only)		2
(g)	20D	BRACKET ASSY	65-45799-65 (used on 65-45799-120 only)		2
(h)	21	BRACKET	Deleted		
	22	BRACKET ASSY	Deleted		
	22	BRACKET ASSY	Deleted		

Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 24)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(i)	23	BRACKET ASSY	65-22486-206		2
	23	BRACKET ASSY	491585		2
(j)	23A	BRACKET ASSY	65-22486-206		1
	23A	BRACKET ASSY	491585		1
	23B	BRACKET, TAI VALVE	65-45799-141		1
	23B	BRACKET, TAI VALVE	65-45799-129 (OPT)		1
(k)	24	BRACKET	65-45799-55		1
	24A	BRACKET, TAI VALVE	65-45799-130		1
(l)	25	BRACKET	65-22486-93		1
	25	BRACKET	463663		1
	25A	BRACKET	65-22486-93 (USED ON 65-45799-148)		1
	25A	BRACKET	463663 (USED ON ON 65-45799-148)		1
(m)	26	BRACKET ASSY	65-45799-25		1
(n)	27	BRACKET	69-42852-1		1
	27	BRACKET	69-42852-800 (OPT)		1
	27A	BOLT	NAS1104-9 (OPT)		3
	27B	NUT	NAS679A4W		3
	27C	BRACKET	65-72085-3 (USED ON 65-45798-9)		1
(o)	28	BRACKET	65-22486-142		1
	28	BRACKET	490549		1
	28	BRACKET	65-45799-95		1
	29	BRACKET	65-22486-141		1
	29	BRACKET	490545		1
	29	BRACKET	65-45799-96		1
(p)	29A	BRACKET ASSY	DELETED		
(q)	29B	BRACKET ASSY	DELETED		
(r)	29C	BRACKET ASSY	DELETED		
(s)	29D	BRACKET	65-22486-137 (USED ON 65-45799-123)		1
	29D	BRACKET	480389 (USED ON 65-45799-123)		1

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(4)(a)	30	BRACKET	65-45799-73		1
(b)	31	BRACKET ASSY	65-45799-65		7
(c)	32	BRACKET ASSY	65-22486-88		1
	32	BRACKET ASSY	466472		1
	32	BRACKET ASSY	65-45799-65		1
(d)	33	BRACKET ASSY	65-45799-65		2
(e)	33A	BRACKET, TAI DUCT	65-45799-131 (USED ON 65-45799-1, -117, -120, -123)		1
	34	BRACKET ASSY	65-45799-65		1
	34A	BRACKET ASSY	65-45799-58		1
(f)	35	BRACKET ASSY	65-45799-35		2
	35	BRACKET ASSY	528841-178		
	35A	BRACKET ASSY	69-71522-1		1
	35B	BRACKET	69-45402-12		1
	35C	BOLT	BACB30NF4-14		2
	35D	NUT	BACN10JC4		3
	35E	WASHER	AN960-416L		2
	35F	BOLT	BACB30NF4-8		1
	35G	WASHER	AN960-416		2
	36	BRACKET ASSY	65-45799-46		1
	37	BRACKET ASSY	65-22486-114		1
	37	BRACKET ASSY	467489		1
(i)	38	BRACKET	65-45799-32		1
	39	BOLT	NAS1104-6		2
	40	RETAINER	BACR10G41		2
	41	RIVET	MS20426D3-5		4
(j)(k)	42	BRACKET ASSY	65-45799-30		3
(l)(m)	42A	BRACKET	65-45799-110		1
	42B	RETAINER	BACR10G32		1
	42C	BOLT	NAS1103-3		1
	42D	RIVET	MS20426D3-(-)		2
(n)	43	BRACKET	65-45799-33		1
	44	BOLT	NAS1104-18		2
	45	RETAINER	BACR10G41		2
	46	RIVET	MS20426D3-5		4

Engine Accessory Support Bracket Installation
Figure 520 (Sheet 26)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(o)	47	BRACKET ASSY	65-22486-87		1
	47	BRACKET ASSY	65-45799-97		1
	47	BRACKET ASSY	466471		1
(p)	48	BRACKET ASSY	65-22486-98		1
	48	BRACKET ASSY	65-45799-88		1
	48	BRACKET ASSY	466472		1
(q)	49	BRACKET ASSY	65-45799-65 (USED ON 65-45799-123)		2
	49	BRACKET ASSY	647557 (USED ON 65-45799-123)		2
(r)	50	RACEWAY	65-64491-2		1
	51	WASHER	AN960-3L		2
	52	SCREW	BACS12CB3-9		4
(s)	53	RACEWAY	69-52264-2		1
(5)(a)	54	BRACKET ASSY	65-45799-76		1
	54	BRACKET ASSY	65-45799-94		1
	55	BOLT	NAS1306-2		2
	56	RETAINER	BACR10G63		2
	57	SCREW	NAS514P440-4		4
	58	NUT	NAS679A04W		4
	58A	NUTPLATE	NAS1067A3		1
	58B	RIVET	MS20426D3-4		2
(b)		REFER TO PARTS STEP (a)			
(c)	59	BRACKET	65-22486-95		2
	59	BRACKET	466630		2
(d)	60	BRACKET	65-22486-148		1
	60	BRACKET	490479		1
(e)	61	BRACKET ASSY	65-45799-22		1
(f)	62	BRACKET	65-22486-230		1
	62	BRACKET	567326		1
(g)	63	BRACKET ASSY	65-45799-39		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(6)(a)	64	BRACKET	65-45799-52 (USED ON 65-45799-1 AND -123)		1
	64	BRACKET	65-45799-118 (USED ON 65-45799-117 AND -120)		1
	64	BRACKET	65-45799-132 (USED ON 65-45799-127)		1
(b)	65	BRACKET ASSY	65-22486-173		1
	65	BRACKET ASSY	491578		1
(c)	66	BRACKET ASSY	65-45799-24		2
	66	BRACKET ASSY	583479		2
(d)	67	BRACKET ASSY	65-45799-28		1
(e)	68	BRACKET	65-45799-34		1
(f)	68A	BRACKET	69-72085-2		1
	68B	BRACKET	69-72085-1		1
(g)	68C	BRACKET	69-66635-1 (USED ON 65-45798-4)		1
	68D	SCREW	BACS12CB3-7		2
	68E	WASHER	AN960D10L		2
(7)(a)	69	BRACKET ASSY	65-45799-36		1
(b)	70	BRACKET ASSY	65-45799-74		1
(c)	71	BRACKET ASSY	65-45799-6		1
	71A	BRACKET	65-45799-8		1
	72	BRACKET	65-45799-9		1
	72A	BOLT	NAS1104-12 (OR EQUIVALENT)		4
	72B	NUT	NAS679A4W (OR EQUIVALENT)		4

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(d)	73	BRACKET	65-45799-82		1
(e)	74	BRACKET	65-22486-50		1
	74	BRACKET	463717		1
(f)	75	BRACKET ASSY	65-45799-40		1
(g)	76	BRACKET ASSY	65-45799-50		1
(h)	77	BRACKET ASSY	65-45799-37		1
(i)	78	BRACKET	65-22486-8		1
	78	BRACKET	463712		1
	78A	BRACKET	65-45799-142		1
(j)	79	FITTING	69-43416-1		2
(k)	80	BRACKET	65-45799-5		1
	80A	BRACKET	65-45799-7		1
	81	BRACKET	65-45799-9		1
	81A	BOLT	NAS1104-12 (OR EQUIVALENT)		4
	81B	NUT	NAS679A4W (OR EQUIVALENT)		4
(l)	82	BRACKET ASSY	65-45799-64		1
(m)	82A	BRACKET ASSY	69-33197-2		1
	82B	SCREW	NAS1801-3-7		3
	82C	WASHER	AN960D10L		3
	82D	NUT	BACN10JC3		3
	82E	BRACKET	69-22486-135		1
(8)(a)	83	BRACKET ASSY	65-45799-27		5
(b)	84	BRACKET ASSY	65-45799-72		1
(c)	85	BRACKET ASSY	65-45799-21		1

Engine Accessory Support Bracket Installation
 Figure 520 (Sheet 28)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(d)	86	BRACKET ASSY	65-45799-27		1
(e)	87	BRACKET	65-45799-99		1
(f)	88	BRACKET	69-41780-1		1
	88A	BOLT	NAS1104-9 (OR EQUIVALENT)		2
	88B	NUT	NAS679A4W (OR EQUIVALENT)		2
	89	BRACKET ASSY	65-45799-47		1
(g)	90	BRACKET	65-45799-100		1
(h)	91	BRACKET	69-52295-2		1
	92	SCREW	BACSL2CB3-6		2
	92A	SCREW	NAS1801-3-14		2
	93	NUT	NAS679A3W		2
	93A	NUT	BACN10JC3		2
	93B	SPACER	NAS43DD3-32		2
(i)	94	BRACKET ASSY	65-45799-47		1
(j)	95	BRACKET ASSY	65-45799-3		1
	96	BRACKET	69-41777-800		1
	96	BRACKET	69-41777-1		1
	96A	BOLT	NAS1304-9 (OR EQUIVALENT)		1
(k)	97	BRACKET	65-22486-16		1
	97	BRACKET	466782		1
	97A	BRACKET ASSY	69-20607-6		1
	98	BRACKET ASSY	65-22486-33		1
	98	BRACKET ASSY	463749		1
(l)	99	BRACKET ASSY	69-45428-10		1
	99	BRACKET ASSY	69-45428-1		1
	100	SCREW	BACSL2CB3-6		2
	101	NUT	NAS679A3W		2
(m)	102	BRACKET ASSY	65-22486-33		2
	102	BRACKET ASSY	463749		2
(n)	102A	BRACKET	69-45402-2		1
	102B	SCREW	BACSL2CB3-6		2
	102C	NUT	NAS679A3W		2
	102D	WASHER	AN960D10L		2

Engine Accessory Support Bracket Installation
Figure 520 (Sheet 29)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(o)	103	BRACKET ASSY	65-22486-33		2
	103A	WASHER	AN960-416		1
(p)	104	BRACKET ASSY	65-45799-15		1
	104	BRACKET	65-45799-109		1
(9)(a)	105	BRACKET ASSY	69-45446-4		1
	105A	BOLT	NAS1104-3		2
	105B	WASHER	AN960D416		2
	105C	NUT	NAS679A4W		2
(b)	106	BRACKET	69-52295-1		1
	106A	NUTPLATE	BACN10MK3-85		1
	106B	RIVET	MS20615-4M		2
	106C	BOLT	NAS1104-3		2
	106D	WASHER	AN960D416		2
	106E	NUT	NAS679A4W		2
(c)	107	BRACKET ASSY	69-45492-3		1
	107A	BOLT	NAS1104-3		2
	107B	WASHER	AN960D416		2
	107C	NUT	NAS679A4W		2
(d)	108	BRACKET ASSY	69-62249-1		1
	108	BRACKET ASSY	69-52283-3		1
	108A	BOLT	NAS1104-3		2
	108B	WASHER	AN960D416		2
	108C	NUT	NAS679A4W		2
(e)	109	BRACKET ASSY	69-45492-2		1
	109A	BOLT	NAS1104-3		3
	109B	WASHER	AN960D416		3
	109C	NUT	NAS679A4W		3
(f)	110	BRACKET	69-45428-14		1
	110	BRACKET	69-45428-4 (OPT)		1
	110A	BOLT	NAS1104-3		2
	110B	WASHER	AN960D416		2
	110C	NUT	NAS679A4W		2
(g)	111	BRACKET	69-43417-1		1
	111A	BOLT	NAS1104-4		2
	111B	WASHER	AN960D416		2
	111C	NUT	NAS679A4W		2

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(10)(a)	112	BRACKET	65-45799-90		1
(b)	113	BRACKET ASSY	65-45799-800		6
	113	BRACKET ASSY	454423		6
(c)	114	BRACKET ASSY	65-45799-137		1
	114	BRACKET ASSY	65-45799-11		1
	114A	BRACKET	65-45799-13		1
	114B	BRACKET	65-45799-135		1
	114B	BRACKET	65-45799-12		1
	114C	BRACKET ASSY	65-45799-138		1
	114C	BRACKET ASSY	65-45799-11		1
	114D	BRACKET	65-45799-13		1
	114E	BRACKET	65-45799-136		1
	114E	BRACKET	65-45799-12		1
	115	BOLT	NAS1104-13 (or equivalent with 65-45799-137 and -138)		6
	115	BOLT	NAS1104-16 (or equivalent with 65-45799-11)		6
	115A	NUT	NAS679A4W (use with NAS1104 bolts)		6
(d)	116	BRACKET	65-45799-71		1
(e)	117	BRACKET	65-45799-105		1
	117	BRACKET	65-45799-19		1
	117A	BRACKET	66-66601-1		1
(f)	118	BRACKET ASSY	65-45799-49		1
	119	BRACKET ASSY	65-45799-77		1
(g)	120	BRACKET ASSY	65-45799-17		1
	121	BRACKET ASSY	65-45799-800		1
	121	BRACKET ASSY	454423		1
(h)	122	BRACKET ASSY	65-45799-800		5
	122	BRACKET ASSY	454423		5

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
(i)	123	BRACKET ASSY	65-22486-86		1	
	123	BRACKET ASSY	463647		1	
	123A	BRACKET ASSY	65-45799-800		1	
	123A	BRACKET ASSY	454423		1	
(j)	124	BRACKET ASSY	65-22486-35		1	
	124	BRACKET ASSY	463649		1	
	124A	BRACKET ASSY	65-45799-800		1	
	124A	BRACKET ASSY	454423		1	
(11)	125	BRACKET	69-43468-1		2	
	(a)	126	BUSHING	69-43487-1		8
		127	BOLT	NAS1104-5		8
	128	NUT	NAS679A4W		8	
	129	WASHER	AN960C416L		AR	
	130	WASHER	AN960C416		8	
(b)	131	RUB STRIP	69-29863-2		1	
	132	BOLT	NAS1105-3		1	
	133	WASHER	BACW10AK-WP5		1	
	134	WASHER	BACW10P16S		1	
	135	NUT	NAS679A5		1	
(c)	136	BRACKET	69-48235-1 (OPT)		1	
	137	BUSHING	65-45798-800 (OPT)		1	
	138	BUSHING	65-45798-801 (OPT)		1	
	139	WASHER, Special	65-45798-802 (OPT)		2	
	140	BOLT	NAS1104-6 (OPT)		2	
	141	NUT	NAS679A4 (OPT)		2	
(12)	142	BRACKET ASSY	65-22486-169		1	
	142	BRACKET ASSY	468904		1	
(13)	143	BRACKET ASSY	65-45799-38		1	
(14)	144	BRACKET	69-25331-8		1	
	144	BRACKET ASSY	69-60439-2		1	
	145	BRACKET ASSY	69-25331-1		1	
	146	BRACKET	69-25331-5		1	
	146	BRACKET ASSY	69-60439-1		1	
	147	BRACKET ASSY	69-25331-2		1	
	148	SCREW	BACSL2CB3-7		6	
	149	WASHER	AN960D10L		6	
	150	NUT	NAS679A3W		2	
(15)	151	BRACKET	65-45799-101		1	
(16)	152	CLAMP ASSY	TA5000BH8AT		1	
	153	SCREW	BACSL2CB3-6		2	
	154	NUT	NAS679A3W		2	

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
(17)	155	BRACKET ASSY	65-45799-57 (USED ON 65-45799-1 AND -123)		1	
	155	BRACKET ASSY	65-45799-133 (USED ON 65-45799-127)		1	
	155A	BRACKET ASSY	65-45799-122 (USED ON 65-45799-117 AND -120)			
	155A	BRACKET ASSY	69-48235-4 (USED ONLY WITH 65-73778 THRUST REVERSER INSTL)			
(18)	156	BRACKET	69-22313-12		1	
	(a)	157	SCREW	BACS12CB3-6		4
		158	NUT	NAS679A3W		4
	(b)	159	BRACKET	69-22313-13		1
	(c)	160	BRACKET ASSY	69-39688-1		1
		161	SCREW	BACS12CB3-7		5
		162	NUT	NAS679A3W		5
		162A	WASHER	AN960D10L		5
	(19)	163	BRACKET	65-45799-802		1
	(20)	164	BRACKET ASSY	69-45402-7		1
(a)		165	NUT	NAS679A4W		3
		166	SPACER	NAS43HT4-11		3
		166A	BRACKET ASSY	69-45402-10		1
		166B	SCREW	BACS12CB3-7		2
		166C	WASHER	AN960-10L		2
		(b)	167	BRACKET	69-53880-1	
167		BRACKET	69-53885-1		1	
168		WASHER	AN960-416		6	
(c)		169	NUTPLATE	BACN10MK3-45		1
		170	SCREW	BACS12CB06-6		2
		171	NUT	NAS679A06W		2
		171A	BRACKET	AN743-BP12		1
		171B	BOLT	BACS12CB3-7		1
		171C	WASHER	AN960PD10L		1
		171D	NUT	BACN10JC3		1
		172	BRACKET ASSY	69-45804-4		1
173		BOLT	NAS1304-2H		2	
174	WASHER	AN960D416		2		

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	174A	BRACKET ASSY	69-45402-9		1
	174B	BOLT	NAS6604H2		2
	174C	WASHER	AN960-416		2
(e)	175	ANGLE	AN743P12		1
	176	BOLT	NAS1303-1H		1
	177	WASHER	AN960D10L		1
(f)	178	BRACKET	65-45799-93		1
(g)	179	BRACKET ASSY	65-45799-102		1
(h)	180	CLAMP ASSY	TA5000BH8AT		1
	181	SCREW	BACS12CB3-6		2
	182	NUT	NAS679A3W		2
(i)	183	BRACKET	65-45799-26		1
(21)	184	BRACKET ASSY	65-45799-20		1
	185	BRACKET ASSY	65-45799-45		1
(22)	186	BRACKET	69-58497-1		1
(a)	186	BRACKET	69-20607-6		1
	187	SCREW	BACS12CB3-7		4
	188	WASHER	AN960D10L		4
	189	NUT	NAS679A3W		4
(b)	190	BRACKET	69-45428-3		1
	190	BRACKET	69-65869-3		1
	191	SCREW	BACS12CB3-8		1
	192	WASHER	AN960D10L		1
	193	NUT	NAS679A3W		1
(c)	194	BRACKET ASSY	69-45428-8		1
	195	SCREW	BACS12CB3-8		1
	196	DELETED	NAS679A3W		
(d)	196A	BRACKET ASSY	69-24228-1		1
	196B	SCREW	NAS1801-3-8		2
	196C	WASHER	AN960D10L		2
	196D	NUT	BACN10JC3		2
(23)	197	BRACKET ASSY	65-45799-124		3
(a)	197	BRACKET ASSY	65-45799-75		3
	197	BRACKET ASSY	516008		3

Engine Accessory Support Bracket Installation

Figure 520 (Sheet 34)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(b)	198	BRACKET ASSY	65-45799-125		1
	198	BRACKET ASSY	65-45799-92		1
	198	BRACKET ASSY	511804		1
(c)	199	BRACKET ASSY	65-45799-801		2
	199	BRACKET ASSY	454425		2
(d)	200	BRACKET ASSY	69-45402-2		1
	201	SCREW	BACS12CB3-6		2
	202	NUT	NAS679A3W		2
	202A	WASHER	AN960D10L		2
(24)	203	BRACKET ASSY	65-45799-80		1
(a)					
(b)	204	BRACKET	65-45799-875		1
(c)	205	BRACKET	65-45799-876		1
(d)	206	BRACKET ASSY	65-45799-801		1
	206	BRACKET ASSY	454425		1
(e)	207	BRACKET	65-45799-4		1
	208	BRACKET	65-45799-10		1
(f)	209	BRACKET ASSY	69-62271-1		1
	209	BRACKET ASSY	69-36634-4		1
	210	SCREW	BACS12CB3-7		4
	211	WASHER	AN960D10L		4
	212	NUT	NAS679A3W		4
	213	BOLT	NAS1104-4		2
	214	WASHER	AN960D416		2
	215	NUT	NAS679A4W		2
(25)	216	SUPPORT ASSY	69-52261-1		1
	216A	SUPPORT ASSY	69-52261-3		1
	217	SCREW	BACS12CB3-6		4
	217A	SCREW	NAS1801-3-14		2
	218	NUT	NAS679A3W		4
	218A	NUT	BACN10JC3		2
	218B	SPACER	NAS43DD3-32		2
(26)	219	SUPPORT ASSY	69-52291-3		1
	219	SUPPORT ASSY	69-52291-1		1
	220	SCREW	BACS12CB3-7		4
	221	WASHER	AN960-10L		4
	222	NUT	NAS679A3W		4

B. Install cone bolts as follows: (Fig. 521)

CAUTION: A LIMITED NUMBER OF CADMIUM PLATED BOLTS BACB30LM12-64 ARE IN USE. FOR GREATER RESISTANCE TO WEAR THEY SHOULD BE REPLACED BY CHROME PLATED BOLTS BACB30GU12-64.

DO NOT LOAD ENGINE FLANGES.

(1) Engine AFT cone bolt installation

(a) 65-19592-3:

- 1) Coat bolts (2) and holes with grease, MIL-G-23827, immediately prior to bolt installation. Install cone bolt (1) between aft engine mount flanges at top centerline of engine. Attach cone bolt with bolt (2), washers (3 and 4), and nut (5). Place washer (3) beneath nut. Install bolt (2) with head forward.
- 2) Tighten nut (5) to allow bolt (2) end play of 0.005-0.010 inch. Install cotter pin (6).
- 3) For 65-19592-3 installation with SB 737-71-1069 or SB 737-71-1289 incorporated, see paragraphs B.(1)(c) and B.(1)(d).

(b) 65-19592-23, -24, -29 and -30 installations:

NOTE: Omit step (b)4) for 65-19592-29, -30 installations.
Omit step (b)6) for 65-19592-23, -29 installations.

CAUTION: NO WASHERS ALLOWED UNDER BOLT HEAD (2A, 5G).

- 1) Coat bolt (2A) and holes with grease, MIL-G-23827, immediately prior to bolt installation, position cone bolt (1) between aft engine mount flanges at top centerline of engine. Install bolt (2A), fittings (5A, 5B), cone bolt (1), washers (3, 4A), and nut (5C). Install up to 3 washers (4A) under nut, minimum of 1 washer is required. Tighten nut (5C) to allow bolt (2A) end play of 0.005-0.010 inch. Install cotter pin (6).
- 2) Place spacer (5D) between aft engine mount flanges and align with holes in fittings (5A, 5B). Use shims (5E, 5F), remove laminations as required, to obtain a gap of 0.000-0.005 inch between spacer (5D) and bushing flange (bushings in engine flange holes are installed by P&W). Apply 2 coats of BMS 10-11, Type 1, primer (F-20.03) to delaminated shims (5E, 5F).

- 3) Coat bolt (5G) and holes with grease, MIL-G-23827, immediately prior to bolt installation. Install bolt (5G) through fittings (5A, 5B), spacer (5D), rub strip (5H), washer (5J, 5K), nut (5L). Install up to 3 washers under nut (5L), minimum of 1 washer is required. Tighten nut (5L) to 660-980 pound-inches and install cotter pin (6).
 - 4) Install cable assembly (5M) using bolt (5N), washer (5P), and nut (5Q).
 - 5) Install retainers (5R, 5S), using screw (5T), washer (5U). Tighten screws and lockwire per 20-50-02, double twist method.
 - 6) Install rub strip (5V), using bolt (5W), washers (5X, 5Y), and nut (5Z).
- (c) 65-19592-3, -23, -24, -29 and -30 installations with SB 737-71-1069 incorporated:

NOTE: Omit step (c)4) for 65-19592-29, -30 installations.

Omit step (c)6) for 65-19592-23, -29 installations.

CAUTION: NO WASHERS ALLOWED UNDER BOLT HEAD (2A, 5G).

- 1) Coat bolt (2A) and holes with grease, MIL-G-23827, immediately prior to bolt installation, position cone bolt (1) between aft engine mount flanges at top centerline of engine. Install bolt (2A), fittings (5A, 5B), cone bolt (1), washers (3, 4A), and nut (5C). Install up to 3 washers (4A) under nut, minimum of one washer is required. Tighten nut (5C) to allow bolt (2A) end play of 0.005-0.010 inch. Install cotter pin (6).
- 2) Place spacer (5D) between aft engine mount flanges and align with holes in fittings (5A, 5B). Use shims (5E, 5F), remove laminations as required, to obtain a gap of 0.000-0.005 inch between spacer (5D) and bushing flange (bushings in engine flange holes are installed by P&W). Apply 2 coats of BMS 10-11, Type 1, primer (F-20.03) to shims (5E, 5F).
- 3) Coat bolt (5G) and holes with grease, MIL-G-23827, immediately prior to bolt installation. Install bolt (5G) through fittings (5A, 5B), spacer (5D), rub strip (5H), washers (5J, 5K) and nut (5L). Tighten nut (5L) to 660-980 inch-pounds. Install cotter pin (6).
- 4) Install retainers (5R, 5S), using screw (5T) and washer (5U). Tighten screws and lockwire per SOPM 20-50-02, double twist method.

- 5) Install rub strip (5V) using bolt (5W), washers (5X, 5Y) and nut (5Z).
- 6) Insert bolt (6G) through washer (6H) and load limiter (6K). Install washers (6J), as required, and one washer (6E) under nut (6D) to provide minimum of 0.06 inch clearance between bolt (6G) and installation nut securing retainer.
- 7) Install load limiter (6K) into side fittings with bolts (6C), nuts (6A) and washers (6B) under nuts.
- 8) Reinstall bolt (6G) with washers (6H, 6J). Use washer (6E) under nut (6D) with crushable spacer (6F).
- 9) Tighten nut (6D) to 65-70 inch-pounds. A minimum of two exposed threads must protrude after tightening.

NOTE: Do not crush crushable spacer while tightening nut (6D).

- (d) 65-19592-3, -23, -24, -29 and -30 installations with SB 737-71-1289 incorporated:

NOTE: Omit step (c)4) for 65-19592-29, -30 installations.

Omit step (c)6) for 65-19592-23, -29 installations.

CAUTION: NO WASHERS ALLOWED UNDER BOLT HEAD (2A, 5G).

- 1) Coat bolt (2A) and holes with grease, MIL-G-23827, immediately prior to bolt installation, position cone bolt (1) between aft engine mount flanges at top centerline of engine. Install bolt (2A), fittings (5A, 5B), cone bolt (1), washers (3, 4A), and nut (5C). Install up to 3 washers (4A) under nut. Tighten nut (5C) to allow bolt (2A) end play of 0.005-0.010 inch. Install cotter pin (6).
- 2) Place spacer (5D) between aft engine mount flanges and align with holes in fittings (5A, 5B). Use shims (5E, 5F), remove laminations as required, to obtain a gap of 0.000-0.005 inch between spacer (5D) and bushing flange (bushings in engine flange holes are installed by P&W). Apply 2 coats of BMS 10-11, Type 1, primer (F-20.03) to shims (5E, 5F).
- 3) Coat bolt (5G) and holes with grease, MIL-G-23827, immediately prior to bolt installation. Install bolt (5G) through fittings (5A, 5B), spacer (5D), rub strip (5H), washers (5J, 5K) and nut (5L). Install up to 2 washers (5K) under nut. Tighten nut (5L) to 660-980 inch-pounds. Install cotter pin (6).

OVERHAUL MANUAL

- 4) Install retainers (5R, 5S), using screw (5T) and washer (5U). Tighten screws and lockwire per SOPM 20-50-02, double twist method.
- 5) Install rub strip (5V) using bolt (5W), washers (5X, 5Y) and nut (5Z).
- 6) Install bolt (6S) through washer assembly (6Q) and load limiter (6T).

NOTE: Steel part of washer assembly mates with bolt head.

- 7) Install load limiter (6T) onto side fittings (5A, 5B) with bolts (6N), nuts (6L) and washers (6M) under nuts. Tighten nuts to 220-410 inch-pounds.
- 8) Reinstall washer assembly (6Q) with chamfered washer (6R) and nut (6P). Tighten nut (6P) until bolt (6S) does not rotate by hand. Then tighten one half turn. Install cotter pin (6U).

NOTE: Washer assembly under bolt head must be seated in countersunk hole.

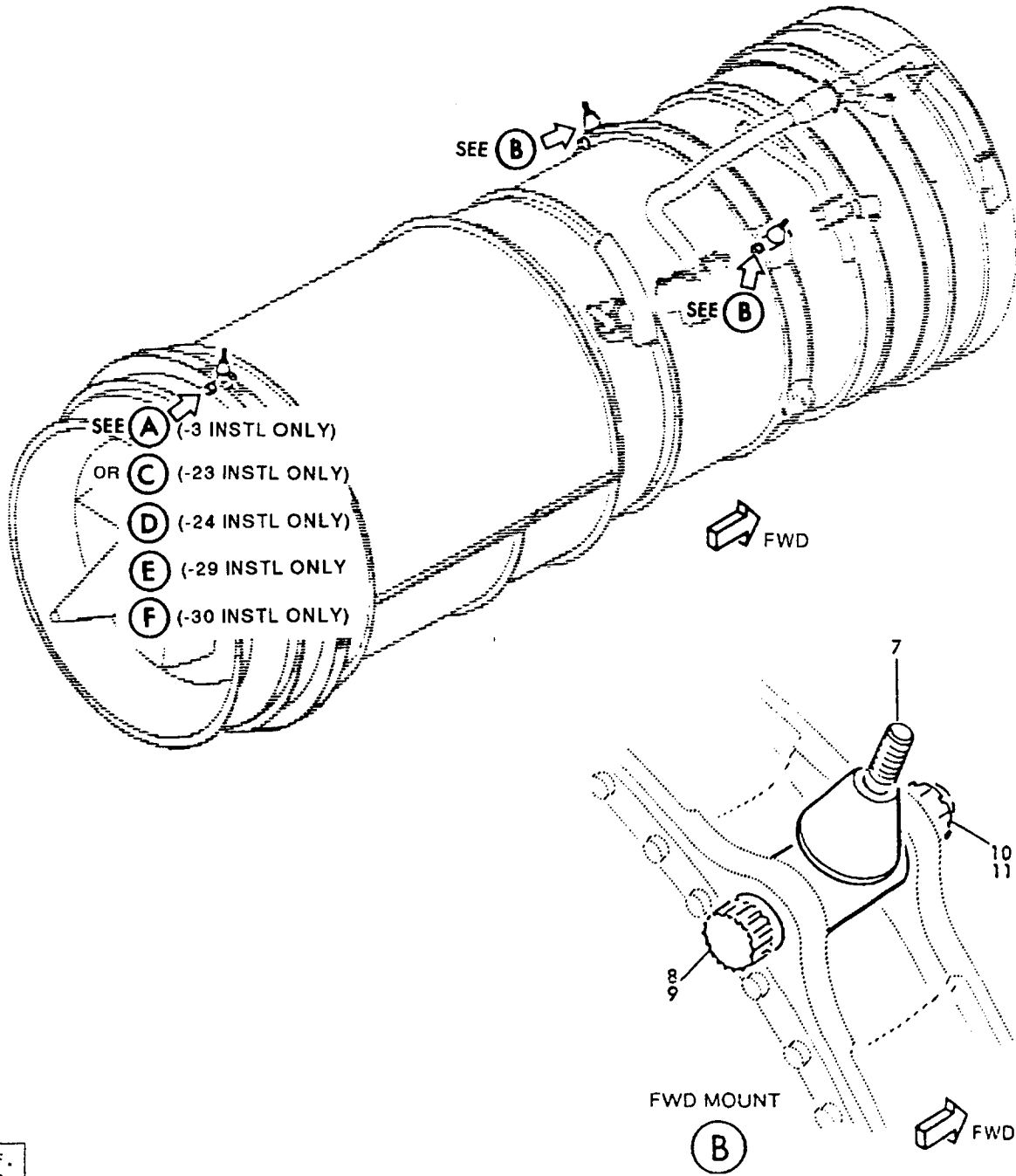
(2) Engine forward cone bolt installation.

- (a) Coat bolts (8) and holes with grease, MIL-G-23827, immediately prior to bolt installation. Locate forward cone bolts (7) between forward engine mount flanges, to left and right of engine centerline. Loosely attach cone bolts with bolts (8), washers (9 and 10), and nuts (11). Install washer (9) with countersunk face against underside of bolthead. Bolthead position, forward or aft, is optional.
- (b) Check per 20-50-01 that locking torque for nut (11) is 90-400 pound-inches for new nut and bolt combination, or 50-400 pound-inches if either nut or bolt is used in subsequent installations.
- (c) Using a spring scale or equivalent, check for free rotation of cone bolt on bolt (8) when a force of less than 10 pounds is applied at cone bolt threaded end.

CAUTION: END PLAY OF CONE BOLT BETWEEN ENGINE FLANGES MUST NOT EXCEED 0.005 INCH OR OVERSTRESS OF ENGINE FLANGES AFTER FINAL TIGHTENING OF BOLT (8) WILL RESULT.

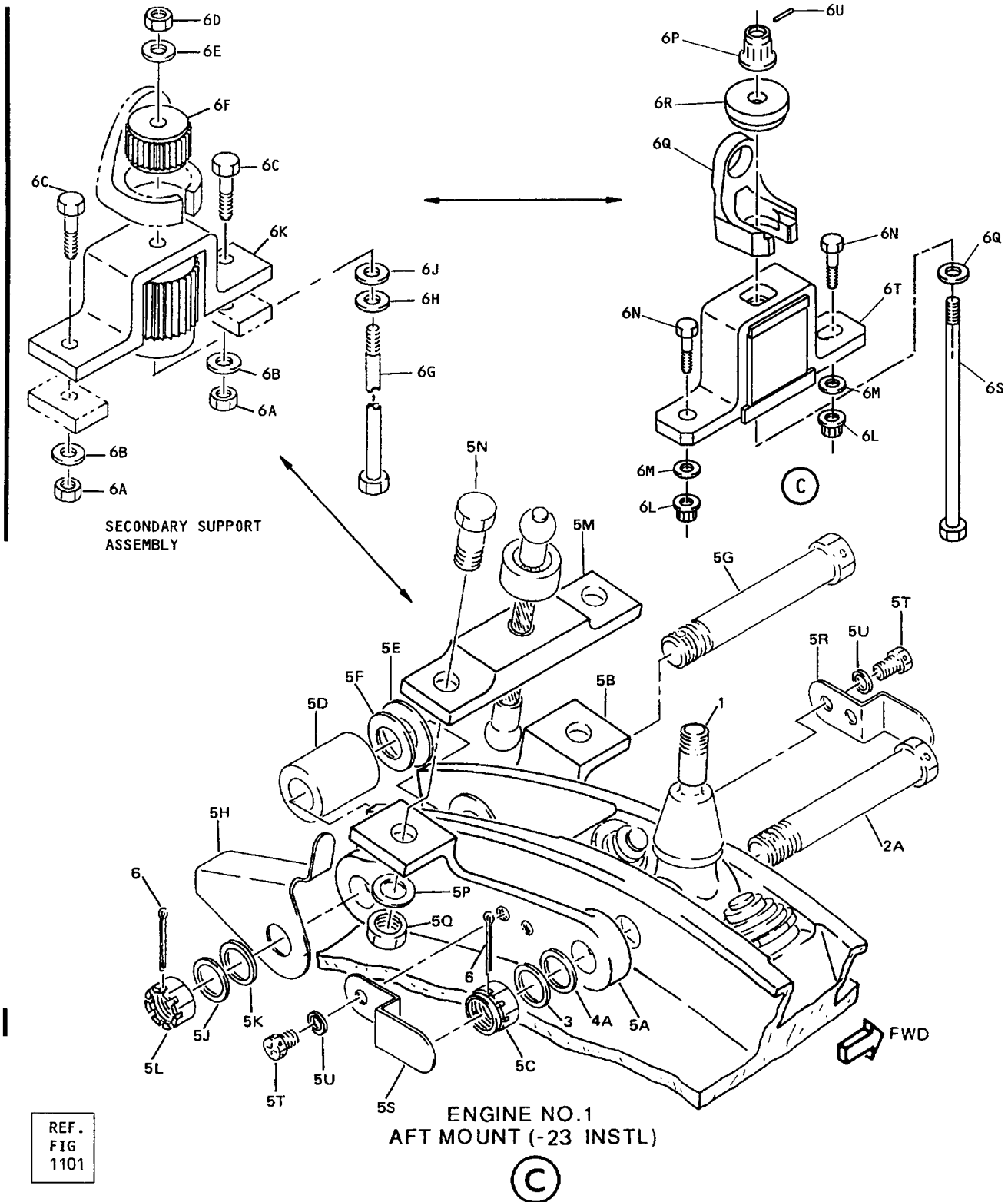
- (d) Check that end play of cone bolt between engine flange lugs does not exceed 0.005 inch total.

- (e) Tighten nuts (11) until a moment of 60 to 75 pound-inches (20 to 25 pounds of force applied to threaded end of bolt) is required to rotate cone bolt about attaching bolt. Final resultant torque of bolt (8) shall not exceed 1000 pound-inches.



REF.
FIG
1101

Engine Mount Cone Bolt Installation
Figure 521 (Sheet 1)



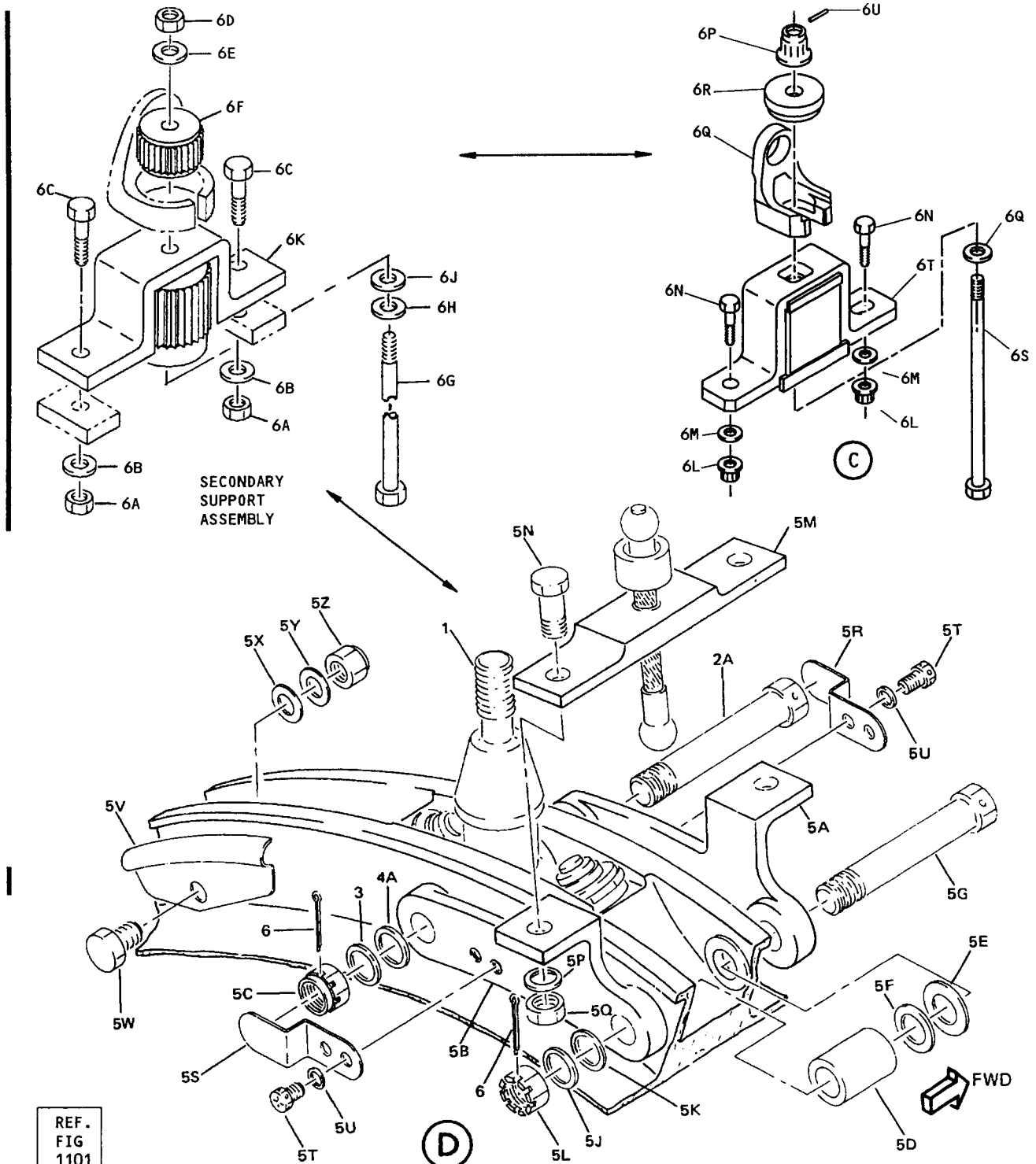
SECONDARY SUPPORT ASSEMBLY

ENGINE NO. 1
 AFT MOUNT (-23 INSTL)

Ⓒ

Engine Mount Cone Bolt Installation
 Figure 521 (Sheet 2)

REF.
 FIG
 1101



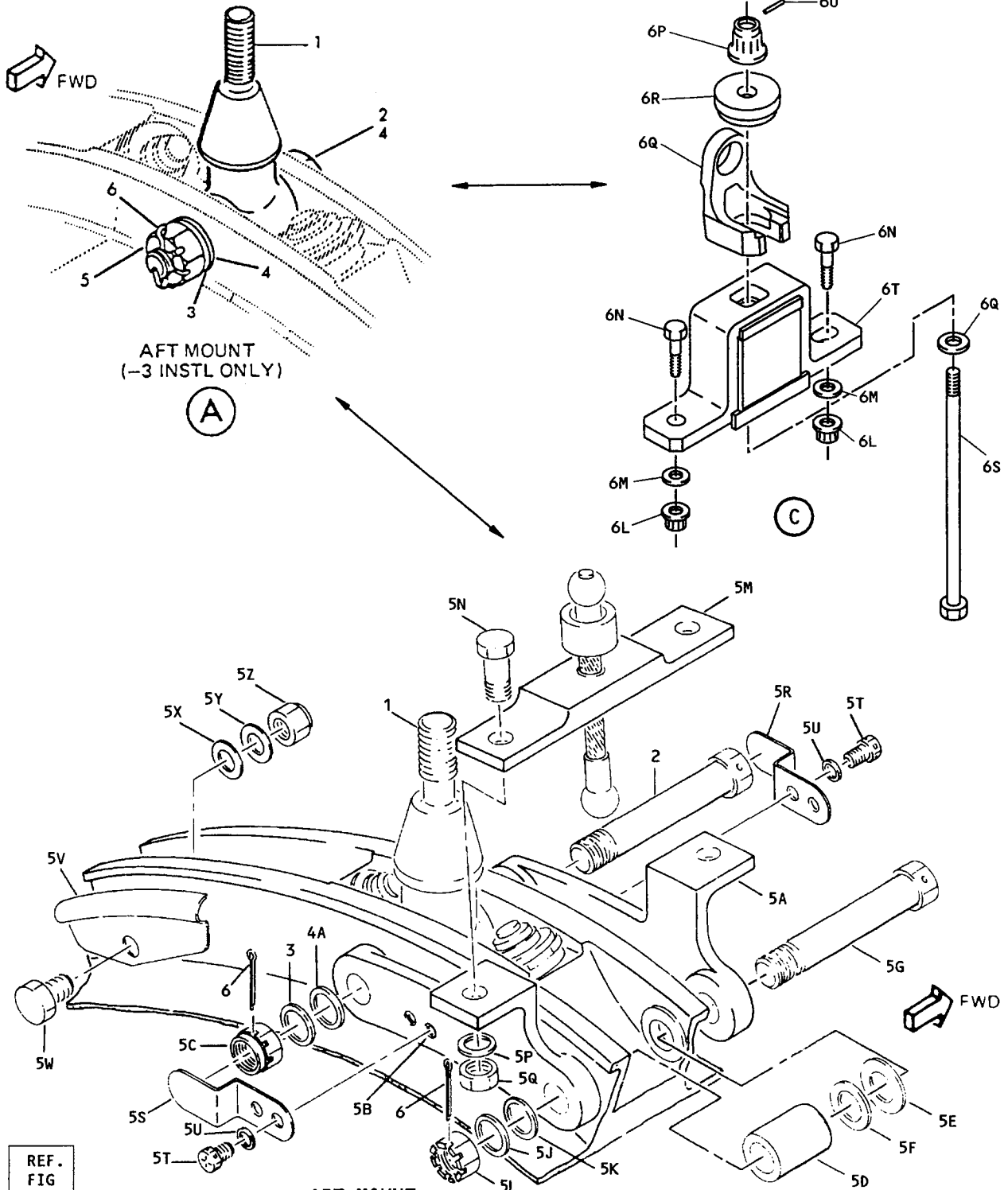
SECONDARY
SUPPORT
ASSEMBLY

REF.
FIG.
1101

ENGINE NO.2
AFT MOUNT (-24 INSTL)
Engine Mount Cone Bolt Installation
Figure 521 (Sheet 2A)

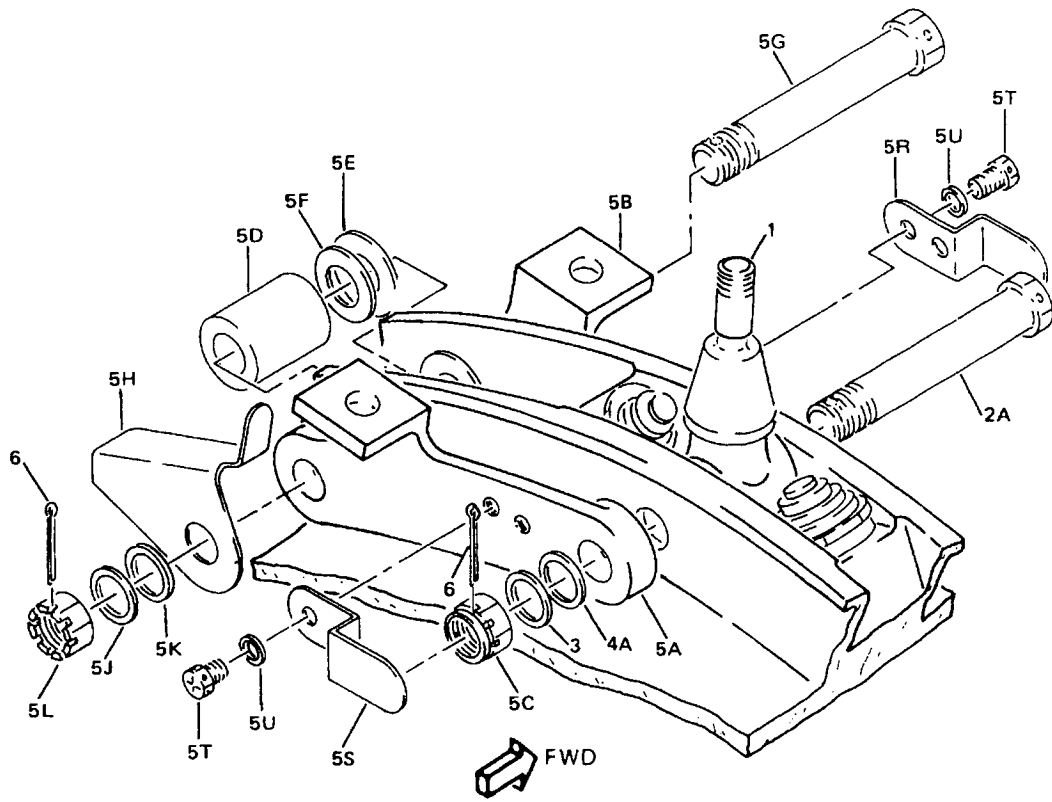
E95444

OVERHAUL MANUAL



REF.
FIG
1101

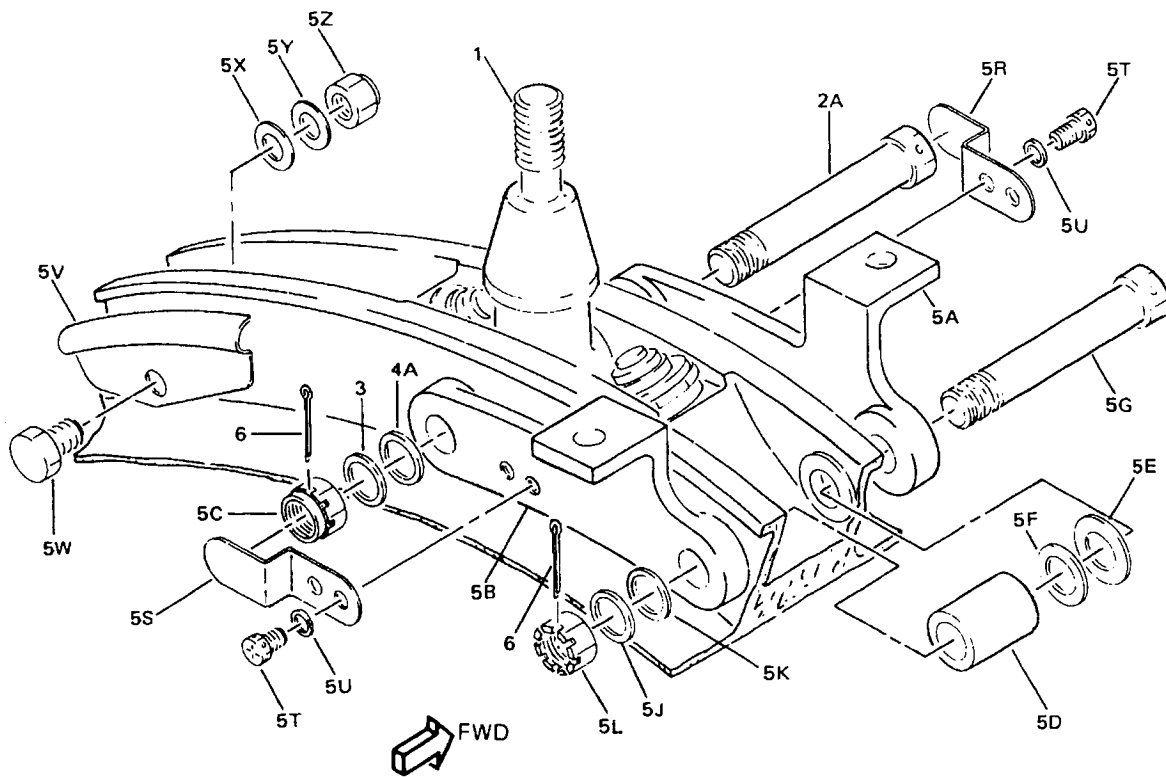
**AFT MOUNT
(-3 INSTL)**
**Engine Mount Cone Bolt Installation
Figure 521 (Sheet 2B)**



ENGINE NO. 1
AFT MOUNT (-29 INSTL)

(E)

Engine Mount Cone Bolt Installation
Figure 521 (Sheet 2C)



ENGINE NO.2
AFT MOUNT (-30 ONLY)

(F)

Engine Mount Cone Bolt Installation
Figure 521 (Sheet 2D)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
B.(1)	(a)1	CONE BOLT, AFT	R18424-1		1	
	1	CONE BOLT, AFT	R18424-2		1	
	2	BOLT	66-13297-2		1	
	2A	BOLT	66-13297-3		1	
	3	WASHER	AN960C916L		1	
	4	WASHER	AN960C916		2	
	4	WASHER	AN960C916		AR	
	5	NUT	AN320C9		1	
	(1)	(b)5A	FITTING	69-72053-1		1
		5A	FITTING	69-72053-3		1
		5B	FITTING	69-72053-2		1
		5B	FITTING	69-72053-4		1
		5C	NUT	BACN10JD109AU		1
		5D	SPACER	65-19592-25		1
		5E	SHIM	69B00412-6		AR
5F		SHIM	69B63073-501		AR	
5G		BOLT	BACB30NF10D51		1	
5H		RUB STRIP	69-72056-2		1	
5H		RUB STRIP	69-72056-4		1	
5J		WASHER	AN9601016L		1	
5K		WASHER	AN9601016		2	
5L		NUT	BACN10JD110AU		1	
5M		CABLE ASSY	69-72057-1		1	
5N		BOLT	NAS6605-8		2	
5P		WASHER	AN960C516		2	
5Q		NUT	NAS1804-5		2	
5R		RETAINER, BOLT	69-72056-1		1	
5S		RETAINER, BOLT	69-72056-3		1	
5T		SCREW	NAS1801-3D6		4	
5U		WASHER	AN960C10L		4	
5V		RUB STRIP	69-29863-2		1	
5W		BOLT	NAS1105-3		1	
5X		WASHER	BACW10AK-WP5		1	
5Y		WASHER	BACW10P-16S		1	
5Z		NUT	BACN10JC5		1	
6		PIN, COTTER	MS24665-370		1	
6A		NUT	NAS1804-5		2	
6B		WASHER	AN960C516		2	
6C		BOLT	NAS6605-9		2	
6D		NUT	84821CD-524		1	
6E		WASHER	AN960C516		1	
6F	SPACER, CRUSHABLE	10-62188-2		1		
6G	BOLT	BACB30LE5-79		1		
6H	WASHER	MS20002C5		1		
6J	WASHER	AN960C516		1		
6K	LIMITER, LOAD	10-62188-1		1		

Engine Mount Cone Bolt Installation
Figure 521 (Sheet 3)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
B.(1)	(b)6L	NUT	NAS1804-6		2
	6M	WASHER	AN960C616		2
	6N	BOLT	NAS6606-10		2
	6P	NUT	BACN10HR5CD		1
	6Q	WASHER ASSY	69-76789-1		1
	6R	WASHER, CHAMFERED	69-76789-4		1
	6S	BOLT	69-78715-1		1
	6T	LIMITER LOAD	65C26824-1		1
	6U	COTTER PIN	MS24665-88		1
	B.(2)	7	CONE BOLT, FWD	R18423-1	
7		CONE BOLT, FWD	R18423-2		2
8		BOLT	BACB30GU12-64		2
8		BOLT	NAS6712E64		2
9		WASHER	MS20002C12		2
10		WASHER	AN960C1216L		4
11		NUT	BACN10JC12		2

Engine Mount Cone Bolt Installation
Figure 521 (Sheet 4)

- C. Install assembly made up of AC generator and constant-speed drive (CSD) as follows: (Fig. 522.)

CAUTION: SUPPORT ENTIRE WEIGHT OF CSD AND GENERATOR ASSEMBLY DURING INSTALLATION.

- (1) Examine CSD pad on engine and adapter plate (2) of QAD assembly (1) to ensure that they are clean and free from burrs and imperfections.
- (2) Lubricate O-ring (3) with Acryloid HF825 lubricant and place in groove in adapter plate (2).
- (3) Lubricate all threaded areas of QAD ring (4) and adapter plate (2) with a 1:1 mixture (by weight) of molybdenum disulphide, Specification MIL-M-7866, and Texaco Unitemp 500 grease.

NOTE: Premixed lubricant Sundstrand P/N 688272-2 (Sundstrand Aviation, Rockford, Illinois) may be substituted.

- (4) Assemble QAD ring and adapter plate.
- (5) Place gasket (5) on engine pad and install adapter plate with bolts (6). Tighten bolts to 180-210 pound-inches.

- (6) Position QAD ring (4) with locking lug just below 3 o'clock position on engine pad.
- (7) Lubricate O-ring (7) with Acryloid HF825 lubricant and place O-ring in groove in CSD case.

CAUTION: IF THERE IS ANY EVIDENCE OF ANTIFRETTING COMPOUND OR GREASE (USED DURING TESTING), CLEAN THE SPLINE WITH STODDARD SOLVENT, SPECIFICATION P-D-680, BEFORE LUBRICATING.

- (8) Lubricate CSD input spline with engine oil.

CAUTION: SUPPORT CSD AND GENERATOR ASSEMBLY IN MANNER TO ENSURE NO WEIGHT ON QAD RING UNTIL RING IS SECURELY SEATED. DO NOT ALLOW UNIT TO HANG ON INPUT SHAFT AND DO NOT BUMP SHAFT DURING INSTALLATION. FAILURE TO SUPPORT UNIT PROPERLY OR BUMPING THE SHAFT CAN DAMAGE PRECISELY FITTED MEMBERS OF INPUT SEAL.

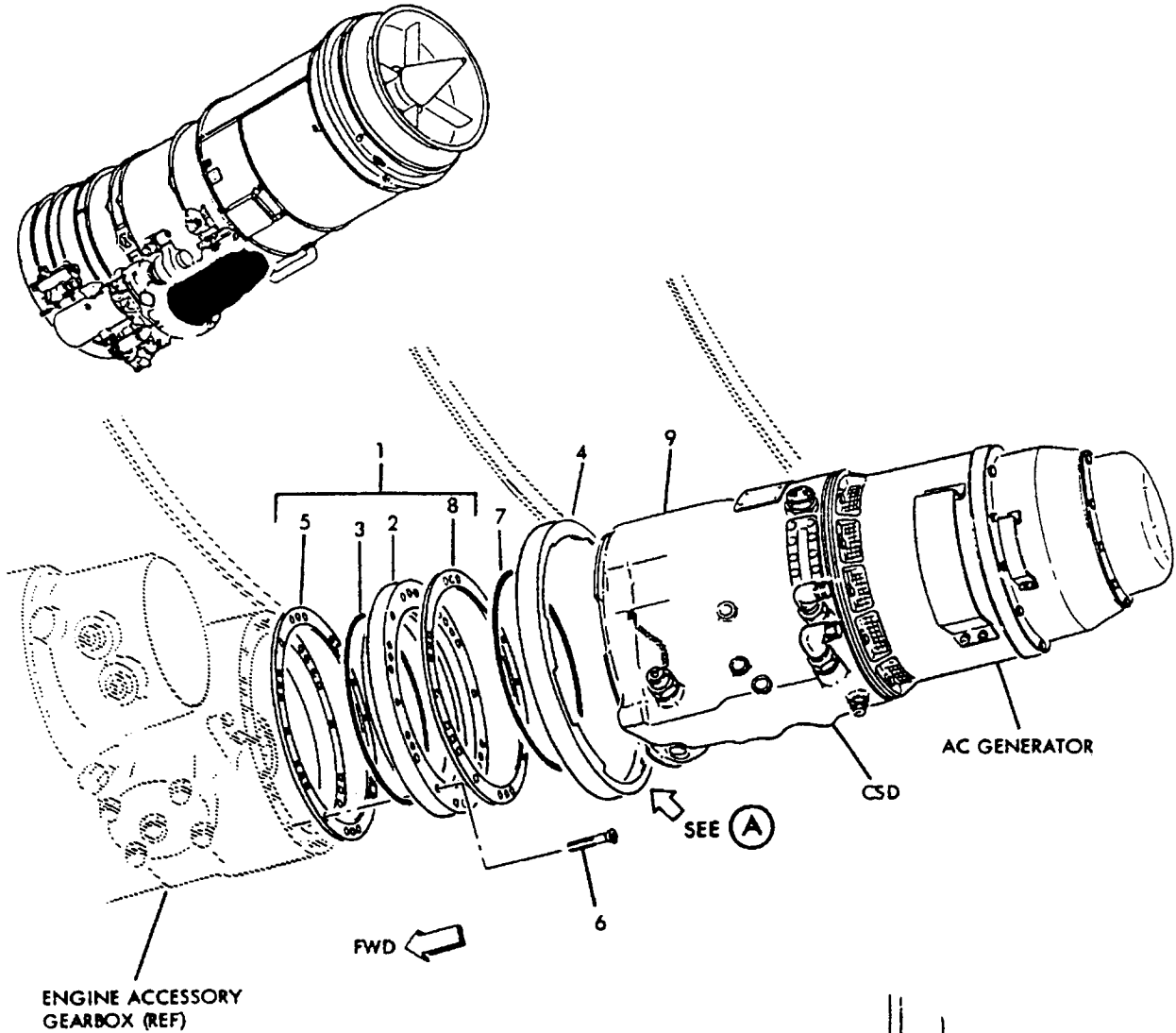
- (9) Place gasket (8) on adapter plate (2) and place CSD and generator assembly (9) in position to be attached, engaging locating pin at 5 o'clock position on CSD with locating hole in QAD adapter.
- (10) Secure CSD and generator assembly to mounting pad as follows:
 - (a) Engage lugs on CSD with lugs on QAD ring.
 - (b) Lubricate universal nut (11) and spherical washer with engine oil, and install nut (11) and lug bolt (10) in QAD ring. Rotate QAD ring while threading bolt through locking lug and nut.
 - (c) Tighten lug bolt (10) 90 to 100 lb-in.
 - (d) Tap exposed outer edge of ring with drift pin of brass or other "soft" material to center ring and eliminate false torque readings.
 - (e) Retighten lug bolt (10) 90 to 100 lb-in. Repeat tapping and tightening until torque reading is not affected by tapping.
 - (f) Check for 3/16 inch minimum clearance between bracket and locking ear. If the dimension is less than 3/16 inch, the input flange should be removed and turned over so that the QAD ring lugs bear on wear-free surfaces.
 - (h) Loosen the lug bolt, without backing off the QAD ring, until a torque value of zero (0) is obtained. Lockwire bolt to QAD ring as shown in Fig. 522 per 20-50-02.

(11) Service CSD spline activity:

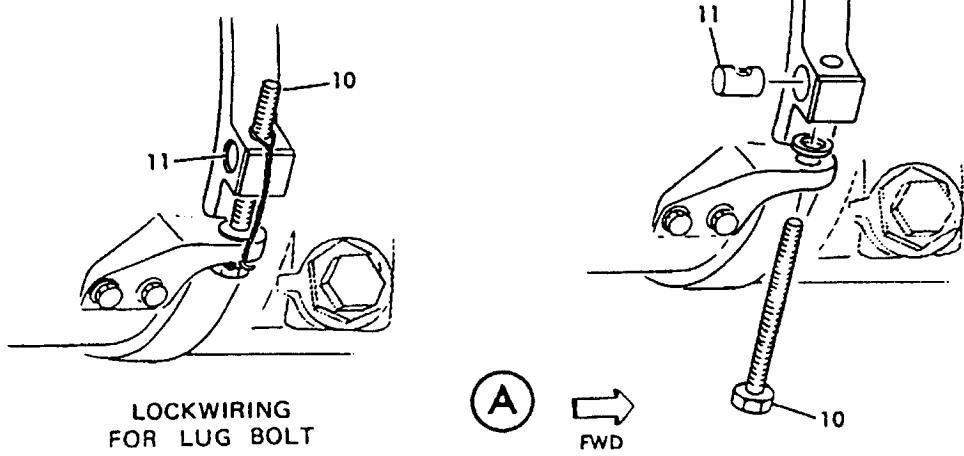
- (a) Remove the cavity standpipe cap on the engine gearbox.
- (b) Remove the spline cavity fill check valve cap from the spline cavity fill check valve.
- (c) Fill the CSD spline cavity through the spline cavity fill check valve to an overflow level with the engine oil conforming to the P&W specification 521B (see P&W Service Bulletin No. 238).

- NOTES:
- 1. Use the pump from the service cart to pump the oil. Approximately 35 psi is required to force the oil through the spline cavity fill check valve.
 - 2. The overflow level is when the oil appears at the opening of the cavity standpipe.

- (d) Disconnect the oil pump connection from the spline cavity fill check valve.
- (e) Reinstall the spline cavity fill check valve cap and tighten to 70-80 lb-in.
- (f) Reinstall the cavity standpipe cap and tighten to 40-50 lb-in.
- (g) Check for leakage between the engine gear case and the constant speed drive case 30 minutes after filling.



REF.
FIG
501
523
1114
1114A



CSD and AC Generator Installation
Figure 522 (Sheet 1)

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
C. (1)	1	QAD ADAPTER ASSEMBLY	10-60295-15 *[1]		1
	2	PLATE, Adapter	Part of 10-60295-15		Ref
(2)	3	O-RING	Part of 10-60295-15		Ref
(3)	4	QAD RING	Part of 10-60295-15		Ref
(4)		NO Parts Required			
(5)	5	GASKET	Part of 10-60295-15		Ref
	6	BOLT	Part of 10-60295-15		Ref
(6)		NO Parts Required			
(7)	7	O-RING	Part of CSD *[1]		Ref
(8)		NO Parts Required			
(9)	8	GASKET	Part of 10-60295-15		Ref
	9	CSD and AC GENERATOR ASSEMBLY	(Preassembled in paragraph 3.B.)		Ref
(10)	10	BOLT, Lug	Part of CSD *[1]		Ref
	11	NUT, Universal	Part of 10-60295-15		Ref
(11)		NO Parts Required			

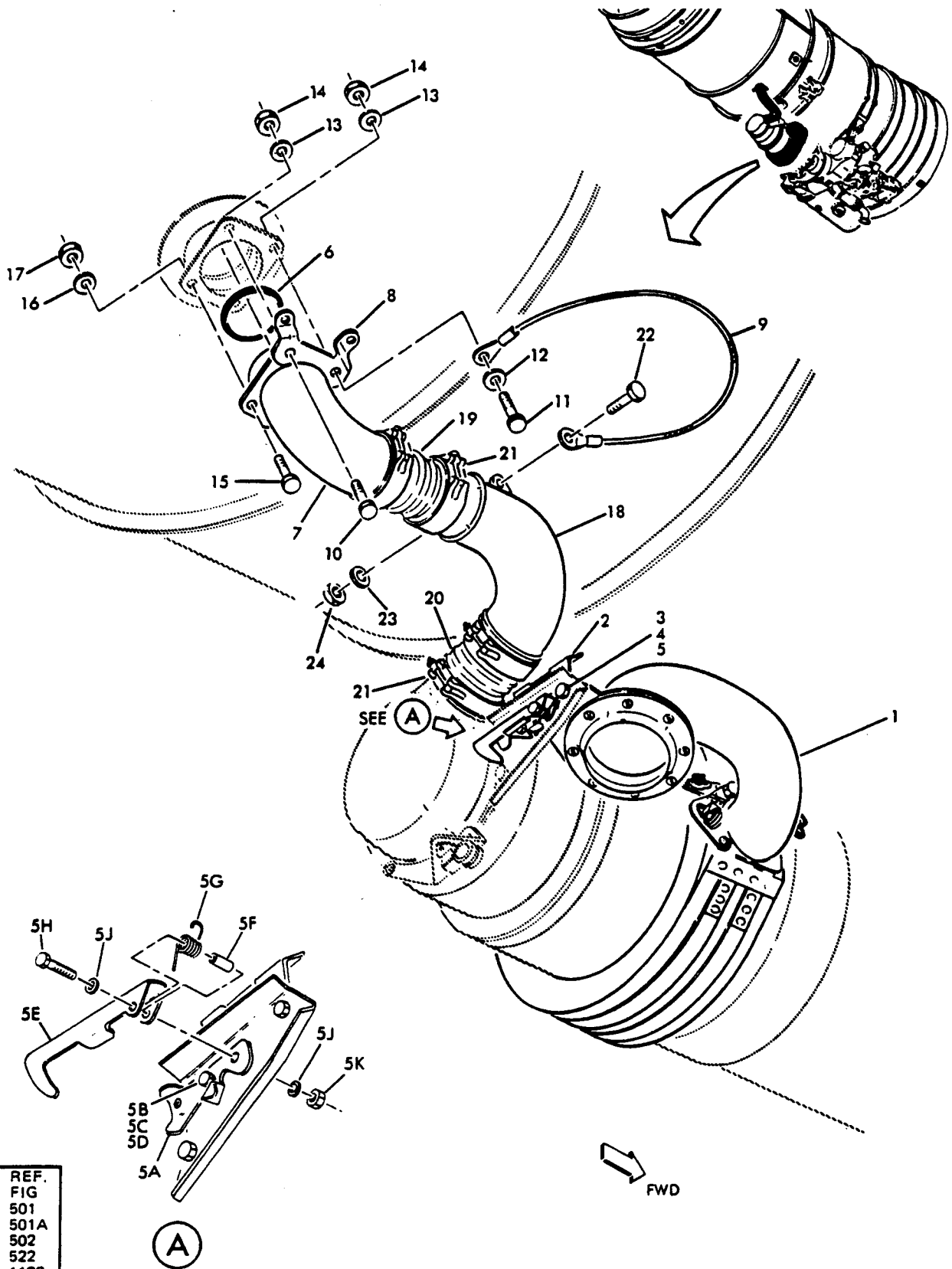
*[1] Refer to Sundstrand Aviation overhaul manual for vendor part number.

737
POWERPLANT BUILDUP

D. Install AC generator cooling ducts as follows: (Fig. 523)

- (1) Install exhaust collector ring assembly (1) as follows:
 - (a) Loosen T-bolt and trunnion halves of clamp on collector ring.
 - (b) Slide collector ring over generator, and locate front flange in groove at front end of generator.
 - (c) Align bolt holes in welded bracket on exhaust duct with bolt holes in duct support bracket on aft end of generator, and tighten T-bolt on collector ring to torque of 30 lb-in.
- (2) Locate bracket assembly (2) and install bolt (3), washer (4) and nut (5).
 - (a) Locate retainer (5A) and install with bolt (5B), washers (5C) and nut (5D).
 - (b) Install hook (5E), spacer (5F) and spring (5G) with bolt (5H), washers (5J) and nut (5K).
- (3) Place O-ring (6) in groove inside cooling air duct (7), and put duct in place over alternator air cooling port in engine case.
- (4) Locate bracket (8) and bonding jumper (9). Attach bracket, bonding jumper, and cooling air duct (7) with bolts (10 and 11), washers (12 and 13), and nuts (14). Tighten nut (14) to 65-70 lb-in.

NOTE: Install washer (12) beneath head of bolt and washer (13) under nut. Make electrical bond between jumper and structure.
- (5) Attach cooling air duct (7) on lower side with bolts (15), washers (16), and nuts (17).
- (6) Install cooling air duct (18) and semiflexible ducts (19 and 20) with clamps (21).
- (7) Attach loose end of bonding jumper (9) to lug on cooling air duct (18) with bolt (22), washer (23), and nut (24). Tighten nut (24) to 65-70 lb-in.



REF.
FIG
501
501A
502
522
1122
1141

AC Generator Cooling Ducts Installation
 Figure 523 (Sheet 1)

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
D. (1)	1	COLLECTOR RING ASSEMBLY	(Preassembled in par. 3.C.)		Ref
(2)	2	BRACKET ASSEMBLY	69-43284-5		1
	3	BOLT	NAS1103-3		1
	4	WASHER	AN960C10L		1
	5	NUT	NAS679A3W		1
	5	NUT	BACN10JC3		1
(2)(a)	5A	RETAINER	69-50910-3		1
	5A	RETAINER	69-50910-5		1
	5B	BOLT	NAS1103-2		1
	5C	WASHER	AN960C10L		2
	5D	NUT	NAS679A3W		1
(2)(b)	5E	HOOK	69-50910-2		1
	5F	SPACER	NAS1057W3-065		1
	5G	SPRING	69-50919-1		1
	5H	BOLT	NAS1103-15		1
	5J	WASHER	AN960C10L		2
	5K	NUT	NAS679A3W		1
(3)	6	O-RING	400922		1
	7	DUCT-COOLING AIR	65-55860-1		1
	7	DUCT-COOLING AIR	65-55860-3		1
(4)	8	BRACKET ASSEMBLY	69-45492-1		1
	9	JUMPER-BONDING	BACJ40A21-8		1
	9	JUMPER-BONDING	BACJ40A21-6		1
	10	BOLT	NAS1104-5		1
	11	BOLT	NAS1104-6		1
	12	WASHER	AN960C416L		1
	13	WASHER	AN960D416L		2
	14	NUT	NAS679A4W		2
	14	NUT	BACN10JC4		2
(5)	15	BOLT	NAS1104-4		2
	16	WASHER	AN960C416L		2
	17	NUT	NAS679A4W		2
	17	NUT	BACN10JC4		2

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(6)	18	DUCT-COOLING AIR	69-42961-1		1
	18	DUCT-COOLING AIR	69-42961-6		1
	19	DUCT-SEMIFLEXIBLE	BACD40D10-15-30		1
	20	DUCT-SEMIFLEXIBLE	BACD40D10-12-30		1
	21	CLAMP	BACC10BN262L-R		4
(7)	22	BOLT	NAS1103-2		1
	23	WASHER	AN960D10		1
	24	NUT	NAS679A3W		1
	24	NUT	BACN10JC3		1

737
POWERPLANT BUILDUP

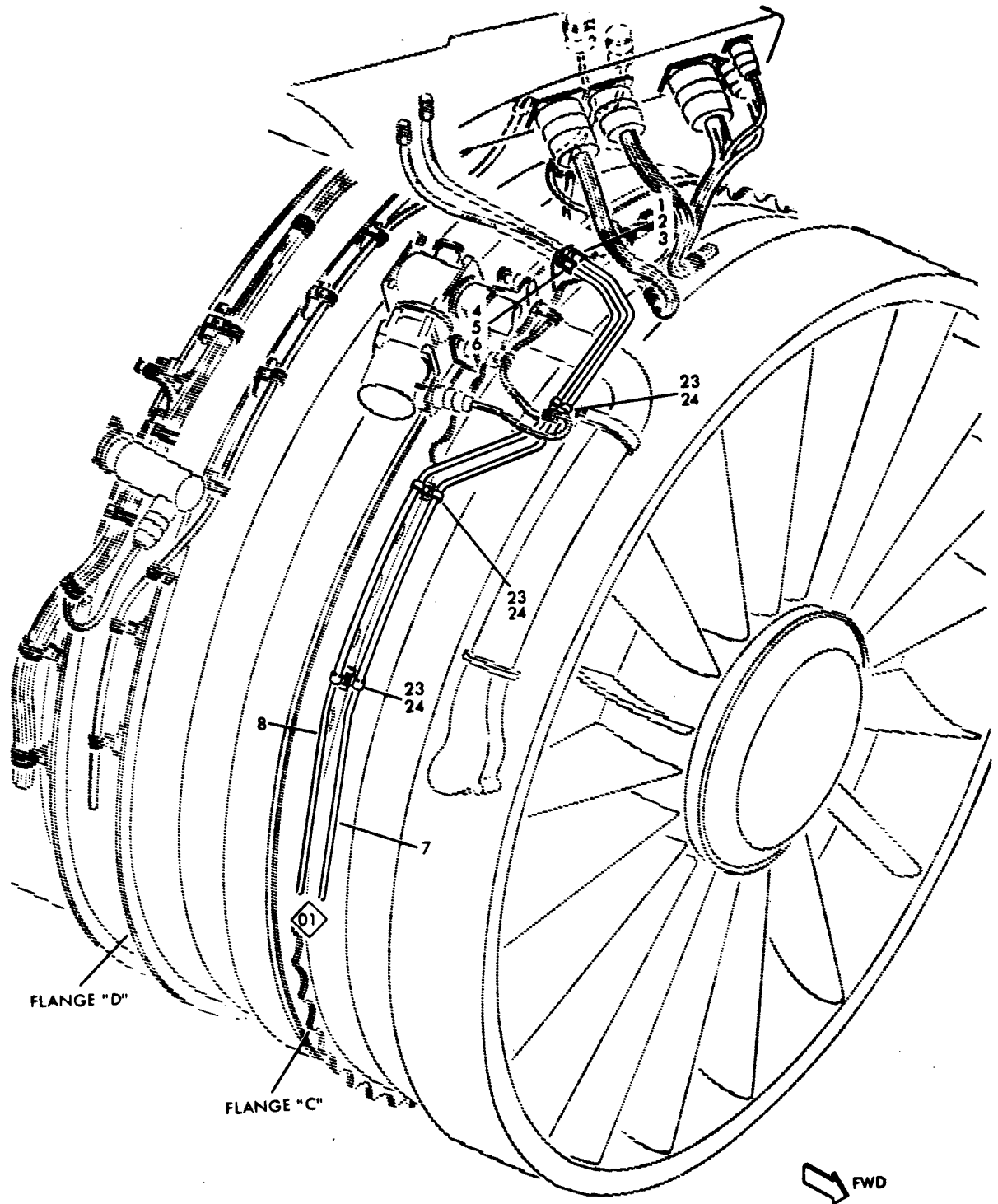
OVERHAUL MANUAL

E. Install pressure ratio plumbing as follows (Fig. 524)

- (1) Install reducer or bulkhead union (1), nut (2), and washer (3) on existing bracket on flange "C" at top of engine. Tighten nut (2).

NOTE: Use nut AN924-6 with union 2-01071 (GA7850 opt) and nut AN924-4 with union BACU24AA0406.

- (2) Install union (4), nut (5), and washer (6) on bracket to right of union (1). Tighten nut (5).
- (3) Attach tube assembly (7) to union (1).
- (4) Attach tube assembly (8) to union (4).
- (5) Install universal elbow (9) in pressure ratio port PT2 at bottom of engine inlet case. Install O-rings (11 and 12) and universal fitting bolt (10). Tighten universal fitting bolt (10) with side port of elbow facing out from engine case. Lockwire bolt in place.
- (6) Attach tube assembly (7) to universal fitting bolt (10). Tighten end fittings of tube assembly.
- (7) Install cap (13) on universal elbow (9).
- (8) Install tee (14), nut (15), and washer (16) on existing bracket on flange "D." Tighten nut (15) with side port of tee facing out from engine.
- (9) Attach tube assembly (8) to forward port of tee (14). Tighten end fittings of tube assembly.
- (10) Install cap (17) on side port of tee (14).
- (11) Attach tube assembly (18) to aft port of tee.
- (12) Install union (19), washer (20), and nut (21) on existing bracket on flange "J." Tighten nut (21).
- (13) Attach tube assembly (18) to union (19). Tighten end fittings of tube assembly.
- (14) Install tube assembly (22) between pressure ratio port PT7 in exhaust case and union (19). Tighten end fittings of tube assembly.
- (15) Install clamps (23) with screws (24) and nuts (25).
- (16) Cap or plug open ports in tubes.



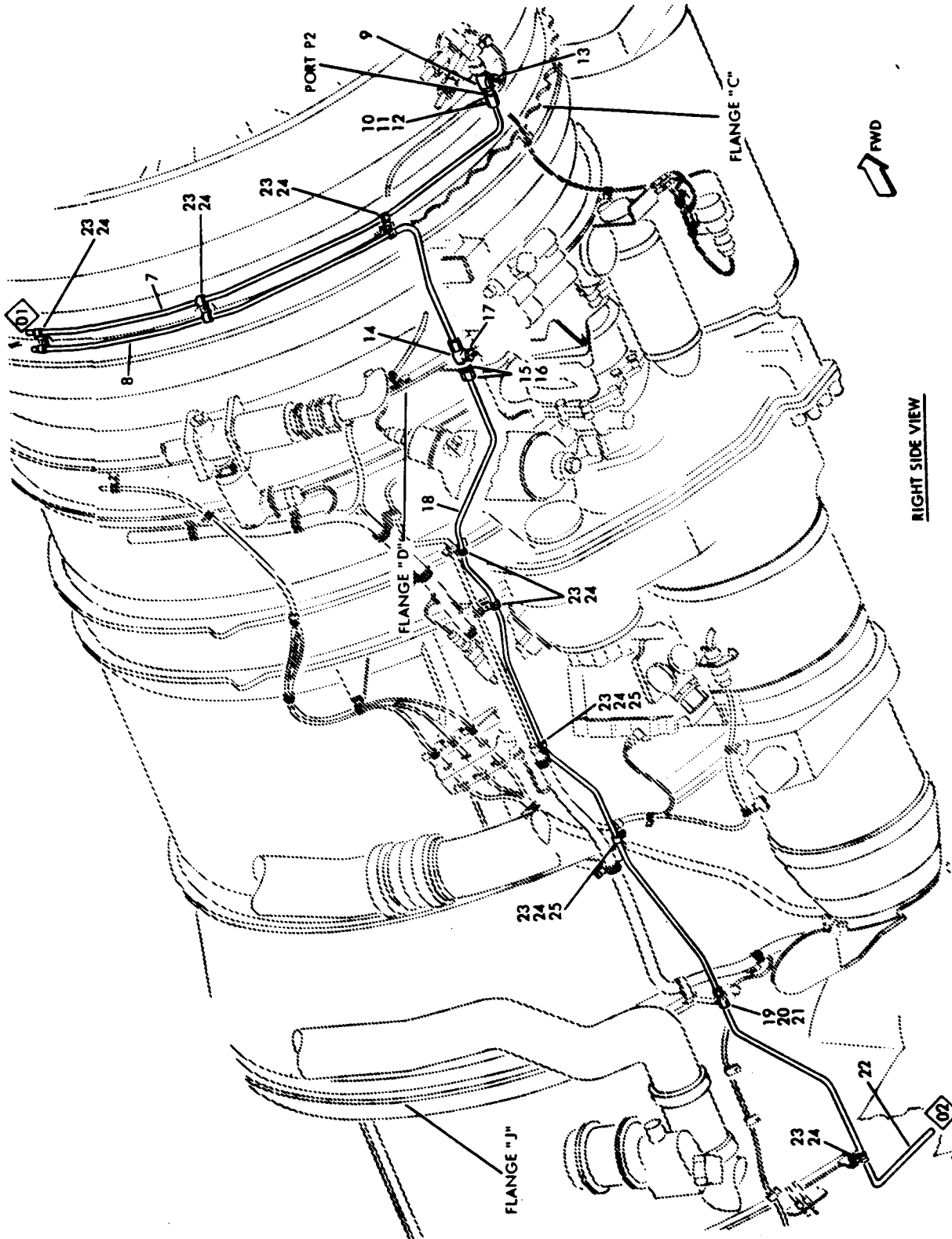
RIGHT SIDE VIEW

REF.
FIG
1126

Pressure Ratio Plumbing
Figure 524 (Sheet 1)

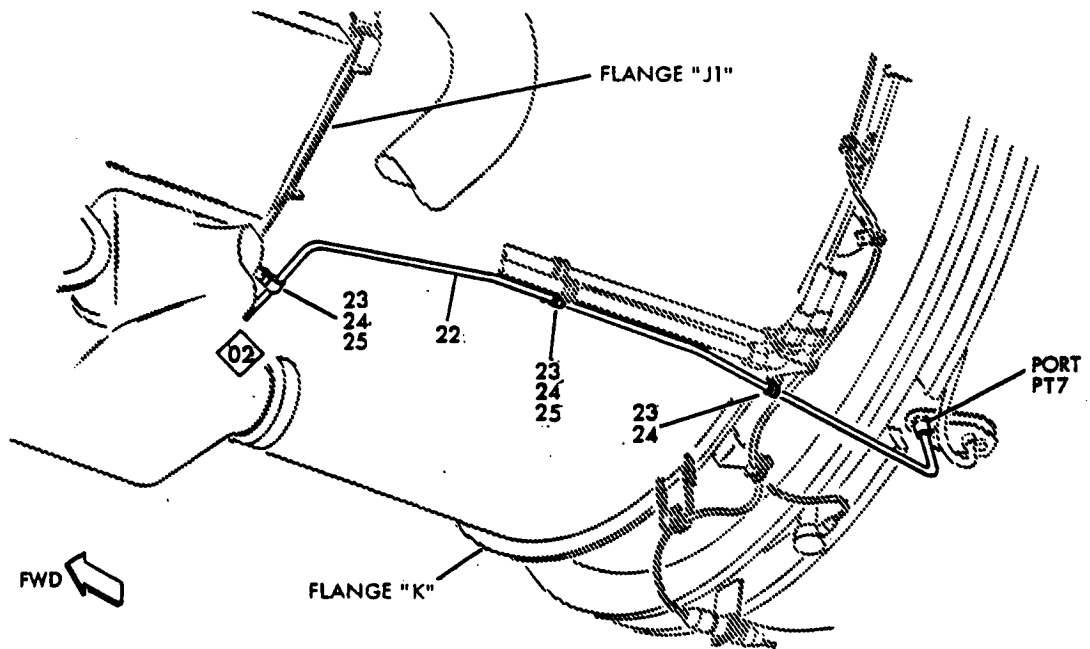
May 15/68

71-03-01
Page 545



Pressure Ratio Plumbing
Figure 524 (Sheet 2)

Aug 15/6i



LEFT SIDE VIEW

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
E.(1)	1	UNION, REDUCER (USE WITH AN924-6)	2-01071		1
	1	UNION, REDUCER (OPT TO 2-01071)	GA7850		1
	1	UNION, BULKHEAD	BACU24AA0406		1
	2	NUT (USE WITH BACU24AA0406)	AN924-4		1
	2	NUT (USE WITH 2-01071)	AN924-6		1
	3	WASHER	AN960-716 AN960-916 (SB 77-1001)		1
(2)	4	UNION	MS21924-4		1
	5	NUT	AN924-4		1
	6	WASHER	AN960-716		1
(3)	7	TUBE ASSY	65-58135-8		1
(4)	8	TUBE ASSY	65-58135-13 65-58135-27 (SB 71-1041, 71-1034, 71-1058)		1

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(5)	9	ELBOW, UNIVERSAL 90	NAS1237-4		1
	10	BOLT, UNIVERSAL FITTING	NAS1236-4H		1
	11	O-RING	BACP11P11		1
	12	O-RING	BACP11P-B4		1
(6)		NO PARTS REQUIRED			
(7)	13	CAP	BACC14H4L		1
(8)		NO PARTS REQUIRED			
(9)	14	TEE	MS21912-4		1
	15	NUT	AN924-4		1
	16	WASHER	AN960-716		1
(10)		NO PARTS REQUIRED			
(11)	17	CAP	BACC14H4L		1
(12)	18	TUBE ASSEMBLY	65-58135-6		1
			65-58135-28 (SB 71-1034, 71-1041)		
			65-58135-29*[1]		
(13)	19	UNION	MS21924-4		1
	20	WASHER	AN960-716		1
	21	NUT	AN924-4		1
(14)		NO PARTS REQUIRED			
(15)	22	TUBE ASSEMBLY	65-58135-5		1
(16)		NO PARTS REQUIRED			
(17)	23	CLAMP	BACC10DD4		22
	24	SCREW	BACS12CB3-7		13
	25	NUT	NAS679A3W		AR
(18)		NO PARTS REQUIRED			

*[1] SB 71-1078, -1099, -1104, -1108, -1138, -1150, -1154, -1159, -1180

F. Install electrical wiring and related components as follows:
(Fig. 525).

(1) General

(a) Refer to 20-11-03, Cleaning Procedures Type V, for information pertaining to proper ground connections. Deviations from wire clamp orientation is permissible when minimum clearances are not otherwise obtainable. Position clamps to maintain adequate clearance and support of wire bundles. The relationship of wire bundle breakout points to adjacent clamps is optional, unless otherwise noted. Determine clamp size and screw and spacer length to suit installation. Provide sufficient slack at each connecting point to prevent strain on connection. Ensure that critical reference points, designated _____, are located correctly.

(b) For steps (4) (k), (m), (r), (s), and (y) following, check electrical resistance between ground connections as follows:

- 1) Maximum resistance between two adjacent wires - .0005 ohms.
- 2) Maximum resistance between top wire and bracket - .001 ohms

CAUTION: DO NOT KINK WIRES, AVOID SHARP BENDS. DO NOT TRAP WIRES. WHEN INSTALLING EQUIPMENT OR TIGHTENING SCREWS AND BOLTS, ENSURE THAT WIRE INSULATION IS NOT BROKEN OR CUT.

- (2) Locate current transformer (1) and bracket (2) on existing bracket between flanges G and H on lower right side of engine. Attach with screws (3), washers (4), and nuts (5). (See view 3, detail A.) Tighten nuts to a torque of 25-35 lb-in.
- (3) Locate heat shield (6) and spacer (7). Attach with screw (8), screws (9), washers (10), and nut (11). (See view 3, detail A.)
- (4) Drape wire bundles (12) and (13) over engine in approximately normal position. Align, attach, and clamp bundles as follows:

NOTE: See view 1 for steps (a) and (b).

(a) Attach terminals of power wires in wire bundle (12) to terminals of generator as follows:

- 1) W200-452-06 to terminal T3
W200-456-06 to terminal T6
W200-453-06 to terminal T2
W200-455-06 to terminal T5
W200-451-06 to terminal T1
W200-454-06 to terminal T4

- 2) Wrap wire bundle, as necessary, to fit the spacer (13L, 13N) with Scotch #64 or TFE grade D, class I Sleeving.

3) Place wire bundle between spacers (13L, 13N). Install screws (13J), washers (13H), and nuts (13F) and tighten nuts (13F).

(b) Attach terminals of voltage regulation wires in wire bundle (13) to generator voltage regulator terminals as follows:

W198-201-18* to terminal A
W198-202-18* to terminal F

NOTE: Refer to appropriate Wiring Diagram Manual for W198-201-18 and W198-202-18 wire color (*).

NOTE: See view 2 and 2A for steps (c), (d), and (e).

(c) Clamp wire bundles (12) and (13) just aft of generator with four clamps (14) and screws (15). Route wire bundle (13) up aft side of flange J and underneath left horizontal flange and down flange K. Install clamps (16, 17, 18, and 19B) with screws (19) and spacer (19C). If vibration detectors are not installed, place connector caps (19D) over connectors D0374 and D0376 (view 7). For installations using pneumatically-operated thrust reverser, attach plug D4824J to existing bracket with four screws (19A); for installations using hydraulically-operated thrust reverser stow free end of wire (par. 6.C. and 7.C.).

(d) Attach one plug D0420 to connector M118 in ignition box or two plugs D1214 and D0420 to connector in ignition box.

(e) Silver braze terminals of wires 457-AL and 457-CH to terminals 1387-4 and 1387-3, respectively, on thermocouple T0202.

NOTE: See view 3 and 3A for steps (f) thru (j).

(f) Route wire bundles (12 and 13) from generator to right side under engine case and attach with clamps (20) and screws (21).

(g) Route fire detector bundle aft from junction with main body of wire bundle (13), and secure with clamps (22) and screws (23). Stow ends of wires.

(h) Connect plug D0006 and lockwire in place. Secure plug wires with clamps (24 and 25), spacer (24A) and screws (26). Stow plug D0766.

(i) Connect plug D0026 from wire bundle (13) to receptacle in current transformer (Detail A).

(j) Route neutral cables W200-454-06, W200-455-06, and W200-456-06 through current transformer (Detail A) and attach with clamps (27) and screws (28).

NOTE: See view 4 for steps (k) thru (o).

737
POWERPLANT BUILDUP

CAUTION: ENSURE THAT NUT (34) DOES NOT BOTTOM OUT BEFORE PROPER CLAMPING OF CABLE TERMINALS TO BRACKET.

- (k) Secure neutral cables with clamps (29) and screws (30). Ground cables on existing bracket on flange "F" with bolt (31), spacer (32), washers (33), and nut (34). Omit one washer as necessary to allow proper installation of nut (34). Tighten nut (34) to 180-200 lb-in. Check maximum electrical resistance per F.(1)(c) (detail B1).

NOTE: If ground wire (35), P/N 61-30202-001 is used, ground wire with neutral cables and perform step (1) (detail B2).

- 1) Secure ground wire (35) along flange "D" with clamps (36) and screws (37).
- (l) Route wire bundle (13) forward and up aft side of flange "E." Secure with clamps (38, 39 and 48) and screws (40). Use spacer (40A) with screw (40B) under one clamp (38) as shown.
- (m) Make ground connections on existing bracket on flange "F" as follows:
- 1) Make ground connection GD14H with terminals of wires W198-160-18, -116-18, -102-18, -129-18.
 - 2) Make ground connection GD14J with terminals of wires W198-122-18, -121-16, -121-18. For engines with vortex dissipator gravel protection system, connect also wire W198-184-18 from the valve and wire W198-181-18 from the pressure switch to GD14J ground connection.
 - 3) Make ground connection GD14K with terminals of wire W198-137-18.
 - 4) Secure grounds with bolts (41), washers (42), and nuts (43) (detail C). Tighten nut (93) to 20-30 lb-in.
- (n) Route fire detector and oil tank unit wire bundles forward and down flange "B." Secure with clamps (44, 45, and 4646) and screws (46A and 47). Fit spacer (47A) to one clamp (45) using screw (47B). Secure one clamp (45) to P&W tube, using clamp (47C) on tube. For engines with vortex dissipator gravel protection system having pressure switch S543 mounted on flange B, install clamp (47D) with existing screw at switch support clamp.
- (o) Connect electrical wiring as follows:
- 1) Connect plug D1006 to receptacle S241.
 - 2) Connect plug D682 to oil quantity indicator system compensator M213.

- 3) For engines with Kidde fire detector system, connect plugs D1092, D1094, D1096 and D1098 to detector element using copper gaskets (47E) in the following manner:

NOTE: Apply light coat of DC-4 grease to copper gasket (47E).

- a) Remove protective cap from electrical connector and check that both connectors are clean and dry.
- b) Fit new copper gasket (47E) into wire bundle connector.
- c) Carefully engage contact pin of bundle connector with detector element receptacle. Tighten bundle connector 50 to 70 pound-inches and lockwire to opposite connector nut.

CAUTION: TO AVOID TWISTING ELEMENT, HOLD ELEMENT CONNECTOR SECURE WHILE TIGHTENING BUNDLE CONNECTOR.

- 4) For engines with vortex dissipator provisions, connect plug D1544 to vortex dissipator valve V87, and connect plug D1546 to pressure switch S543.

NOTE: See view 5 for steps (p) thru (v).

- (p) Route wire bundle (13) up along flange E. Secure with clamps (48, 49) and screws (50).
- (q) Mate plug D1004 with receptacle in valve V57.
- (r) Make dc ground connection GD14F with wires W198-112-18, -120-18, -127-18, and -107-18. Attach wire W198-108-18 to the dc ground connection if the power source is 28 vdc. Secure on existing bracket on flange "F" with bolt (51), nut (52) and washer (52A). Tighten nut (52) to 20 to 30 lb-in.
- (s) Make ac ground connection GD14E with wires W198-124-18, -113-18, and -128-18. Attach wire W198-108-18 to the ac ground connection if the power source is 115 vac. On the alternate configuration, make the ac connection GD14G with wire W198-190-18. Secure on the existing bracket on flange "F" with bolt (51), nut (52) and washer (52A). Tighten nut (52) to 20 to 30 lb-in.
- (t) Install clamps (53) and screws (54).
- (u) Mate plug D0198 to connector on valve V5.
- (v) Install clamps (55), screws (56), spacer (56F), and tube clamp (56G).

NOTE: See view 6 for steps (w) thru (aa).

- (w) Start at generator and route wire bundle (12) forward and up left side. Install clamps (57) with screws (58).

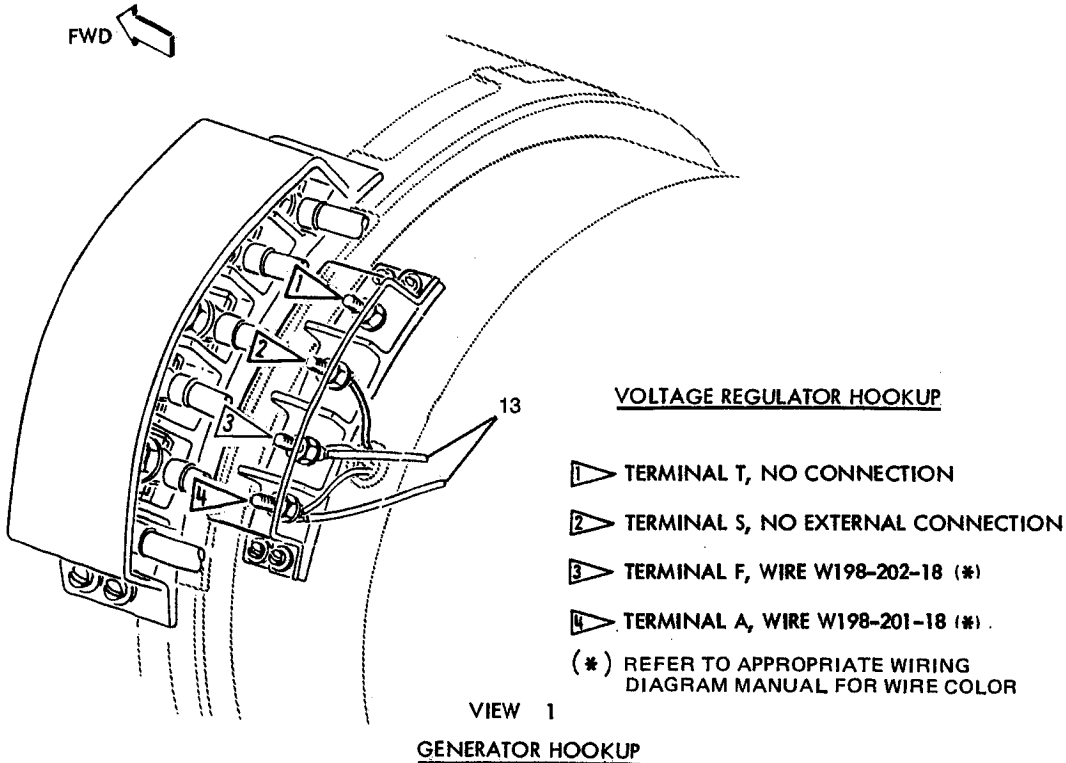
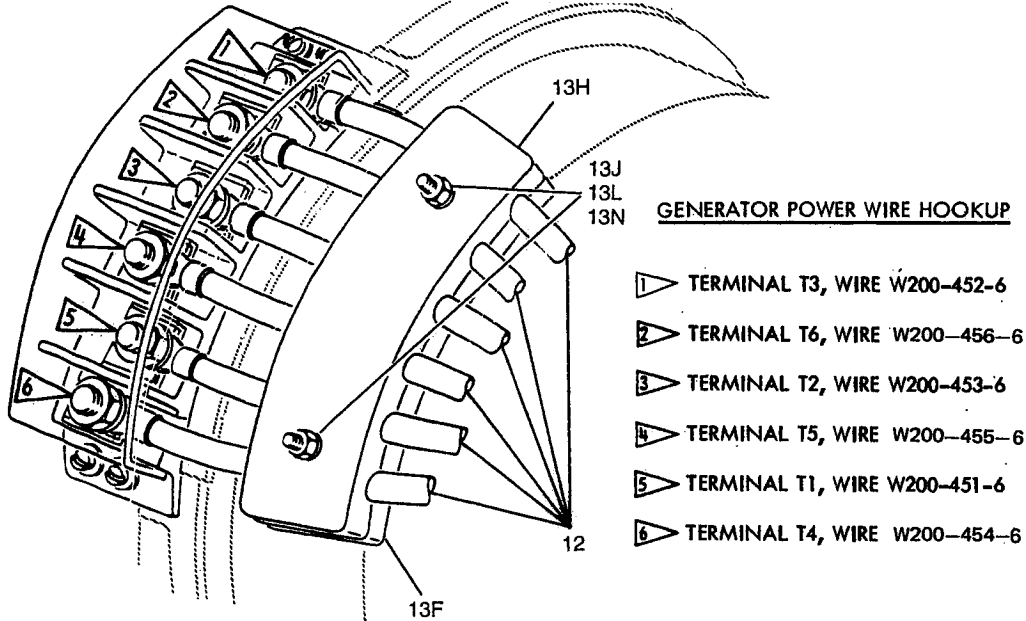
- (x) Secure wire bundle (13), starting at engine starter pad, with clamps (59, 60, 61A, and 61B) screws (61) and spacers (61C, 61D).
- (y) Make ground connections on bonding bracket on flange F as follows:
 - 1) Make dc ground connection GD14D with terminals of wires W198-114-18, -115-18, -117-18, and -133-18.
 - 2) Make ac ground connection GD14C with terminals of wires W198-103-18, -016-18, -131-18, and -135-18.
 - 3) Secure terminals with screws (62), nuts (63) and washers (63A). Tighten nuts (63) to 20-30 lb-in.
- (z) Route wire bundles (12 and 13) up and forward and secure with clamps (64, 65, 66, 67 and 68A) and screws (68 and 68B). In alternate routing, one double clamping position on flange E requires spacer (68C) and screw (68D).
- (aa) Route fuel flowmeter wire bundle down aft side of flange B and secure with clamps (69) and screws (70).

NOTE: See view 7 for steps (ab) and (ac).
- (ab) Route wire bundle (12) up forward side of flange C and wire bundle (13) up aft side of flange B. Secure bundles with clamps (71, 71A, 72, 73, and 74), screws (75), spacer (75A), and screw (75B). For alternate configuration, route wire bundle (12) up between flanges D and E clamping to one or the other as shown in view 7 alternate with clamps (73) and screws (75). Route wire bundle (13) up forward side of flange B with clamps (74) and screws (75) located as shown.
- (ac) Mate plug D0194 to connector on valve V3.

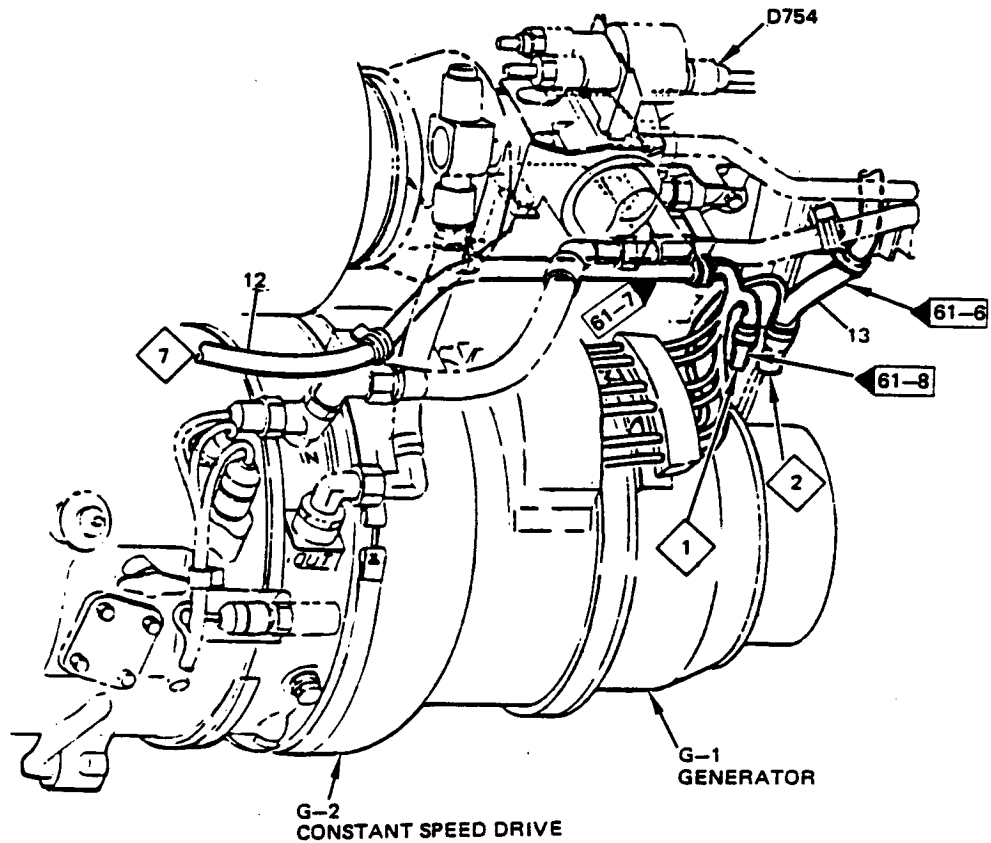
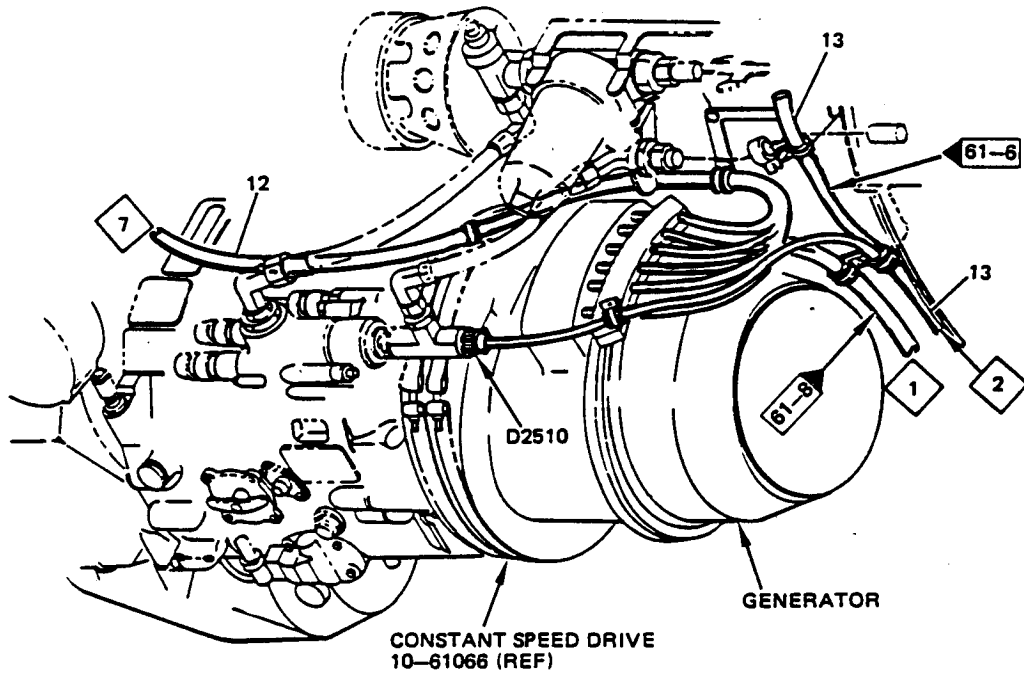
NOTE: See view 8 for steps (ad) through (af).
- (ad) Route wire bundle (76) through inlet valve to accessory flange. Attach with clamp (77), screw (78), and nut (79). Tighten nut (79) to 25-35 lb-in.
- (ae) Connect wire bundle (76) with wire bundle (13) with splices (80). Stamp or mark "SP002" and "SP004" on sleeves. Match sleeves to splice positions as shown in Fig. 525 and install sleeves over splices.

NOTE: Install sleeve so that 1.5 inches of sleeve will extend inside port of inlet accessory case.
- (af) Install clamp (81) with screw (82).

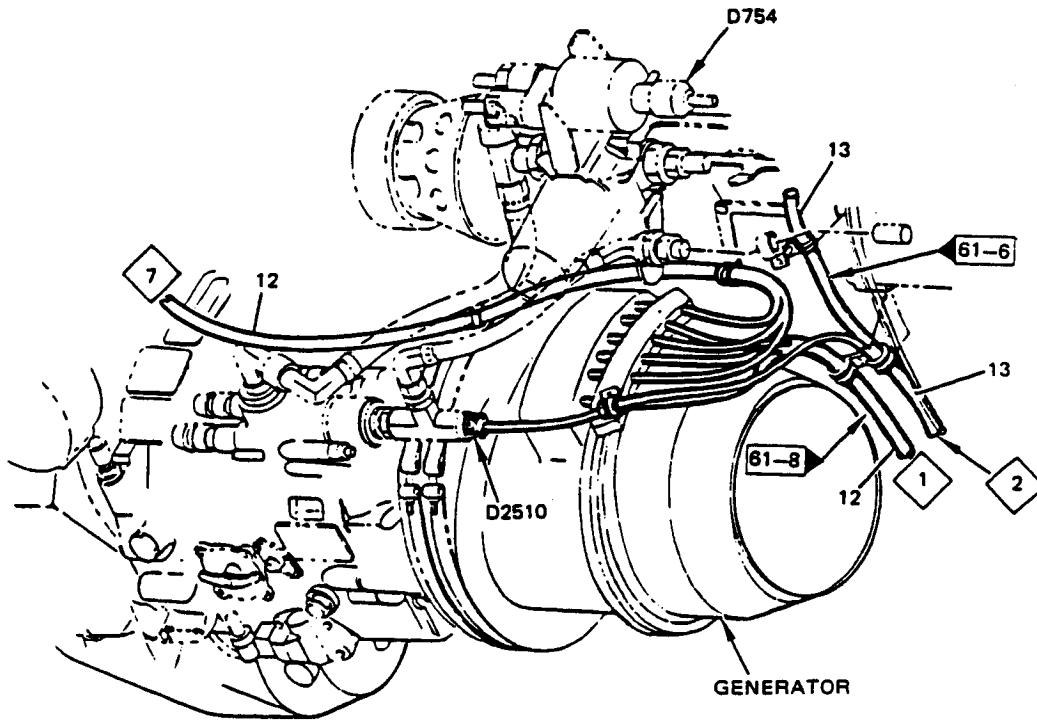
(ag) Route starter valve pressure switch wiring and secure in place using clamp (90), washer (92), spacer (94), screws (96, 98), and nut (100). Install connector furnished with pressure switch to D1864 located on flange G. Ensure that ground strap is installed on P&W bracket mounted at this position.



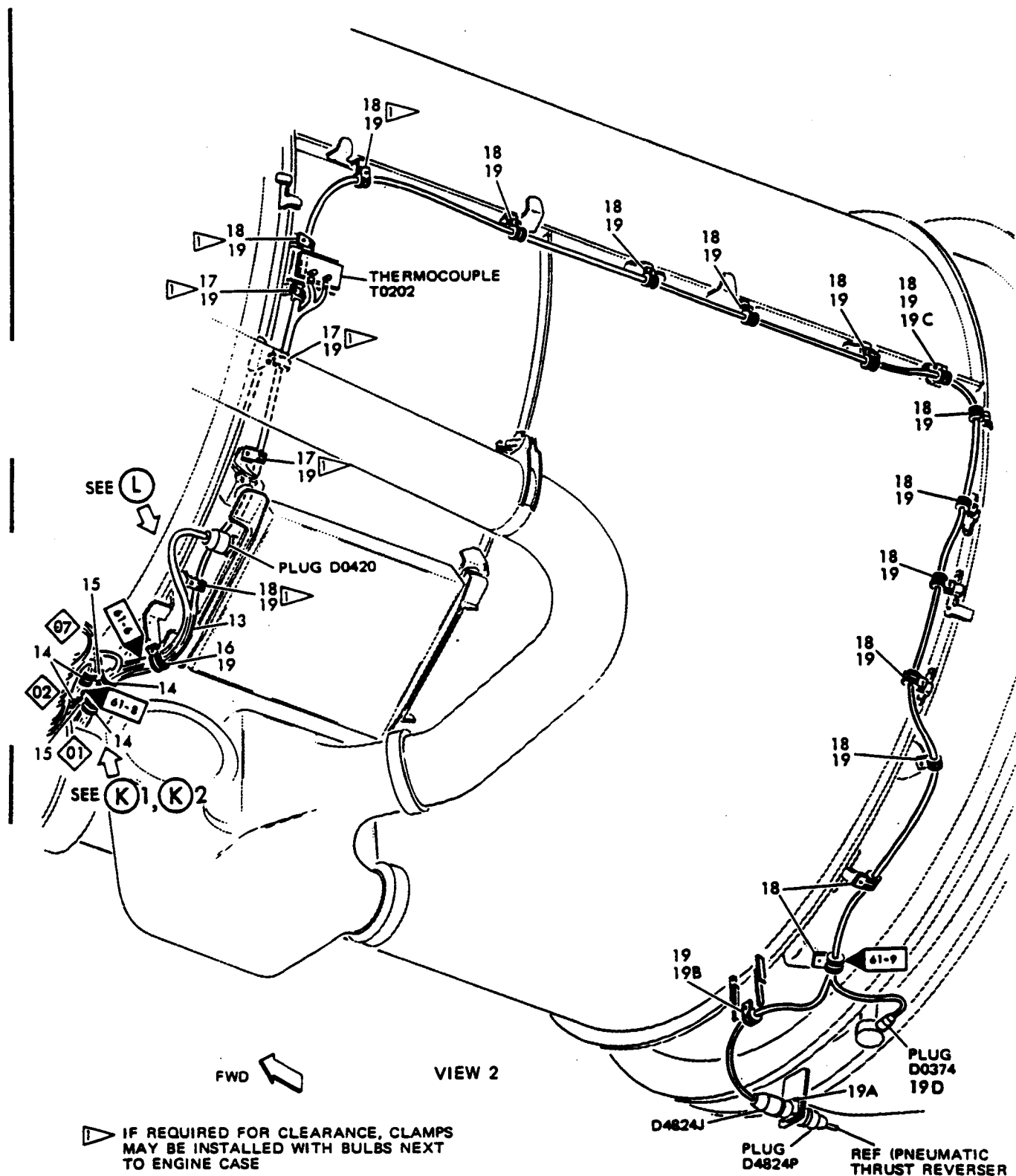
REF.
 FIG
 1117



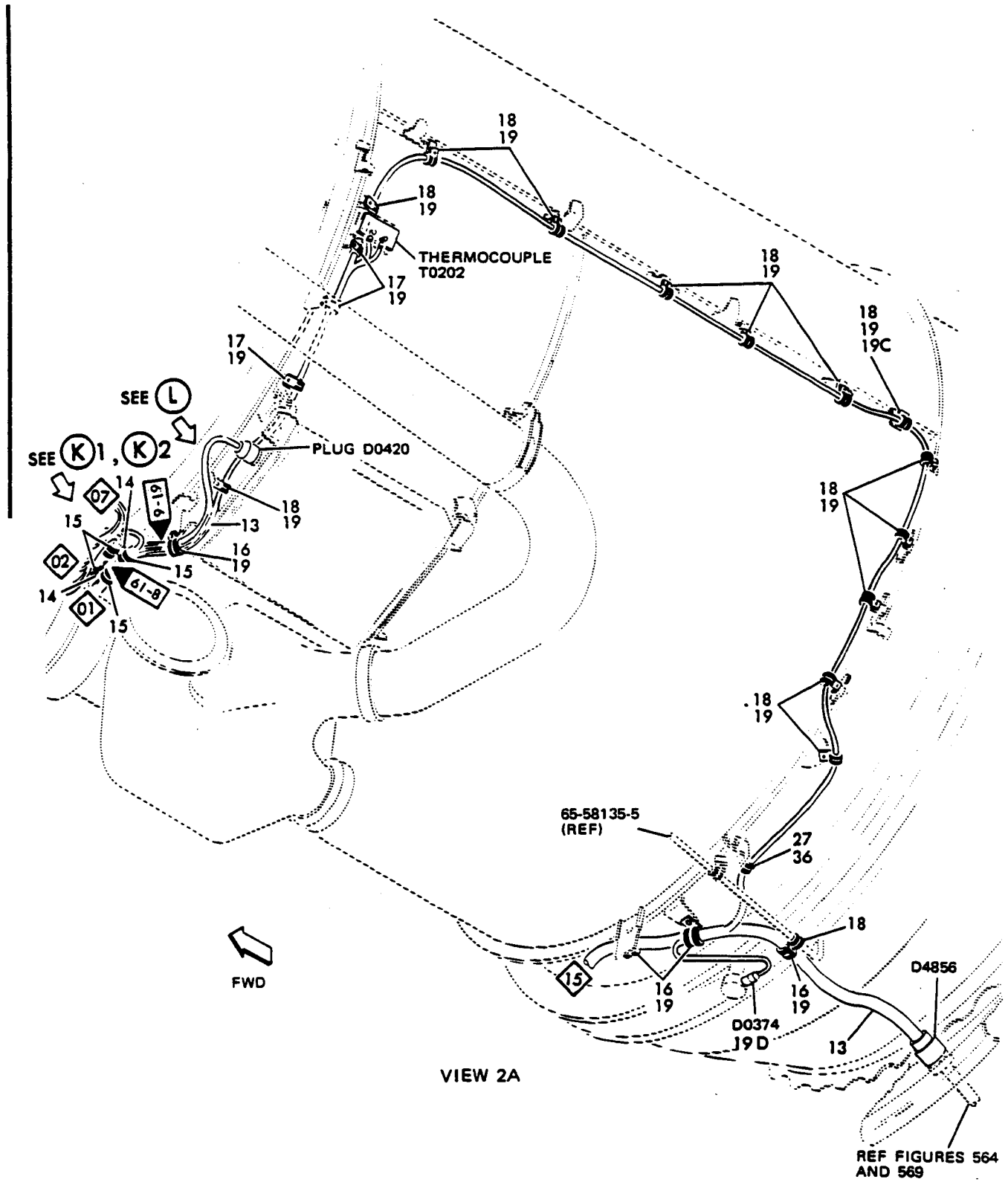
Electrical Installations
Figure 525 (Sheet 2)



(K) 2



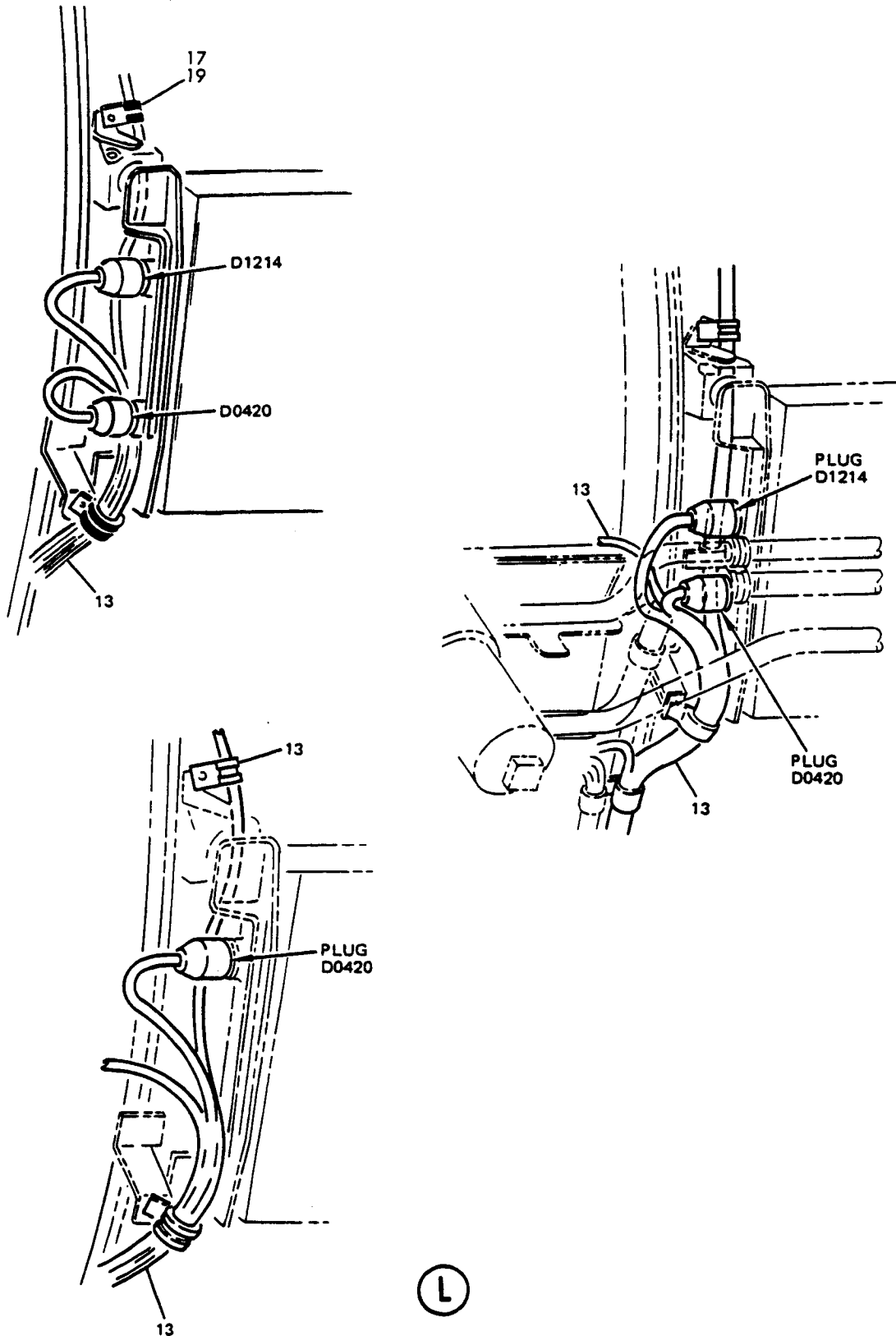
OVERHAUL MANUAL



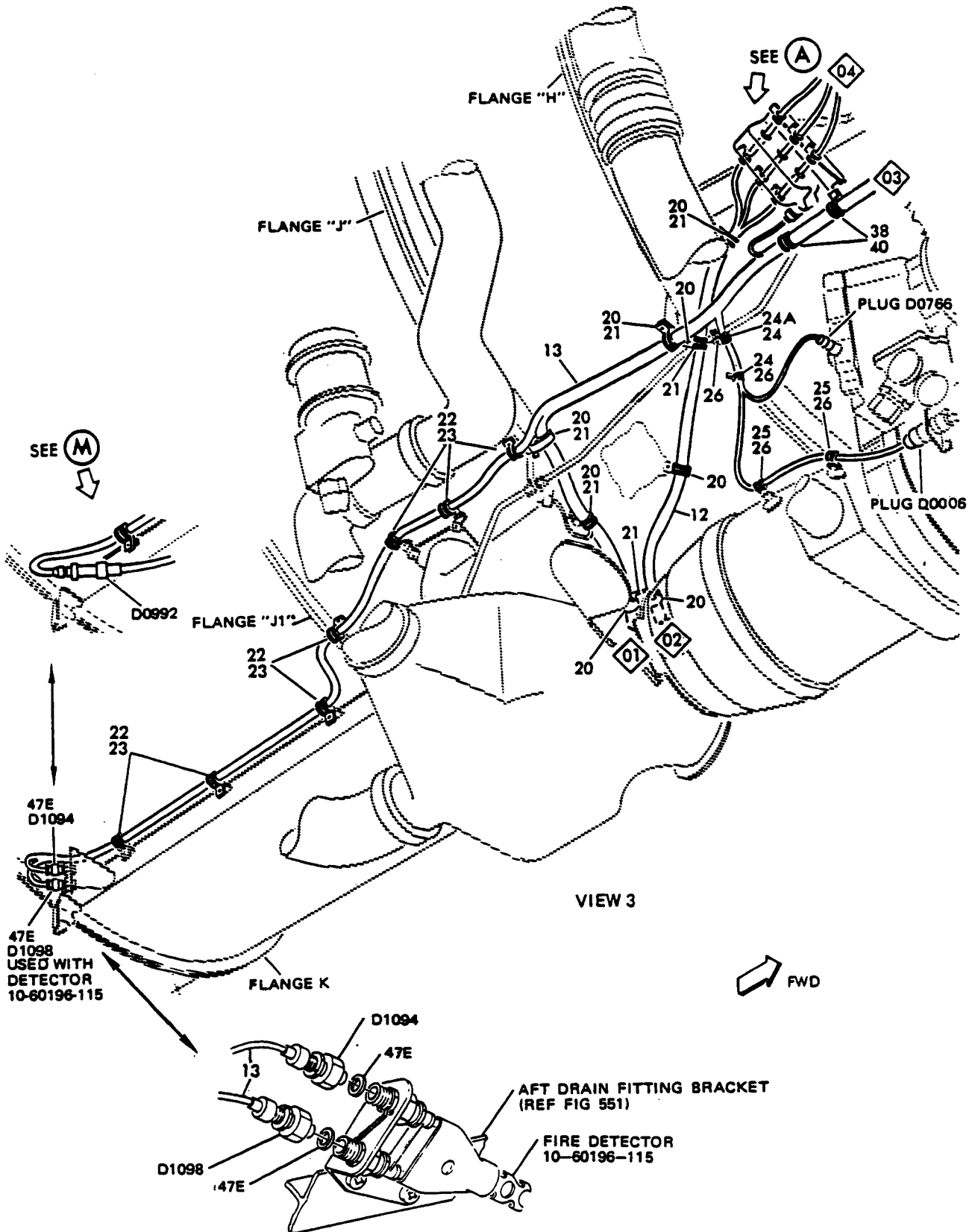
VIEW 2A

Electrical Installations
 Figure 525 (Sheet 5)

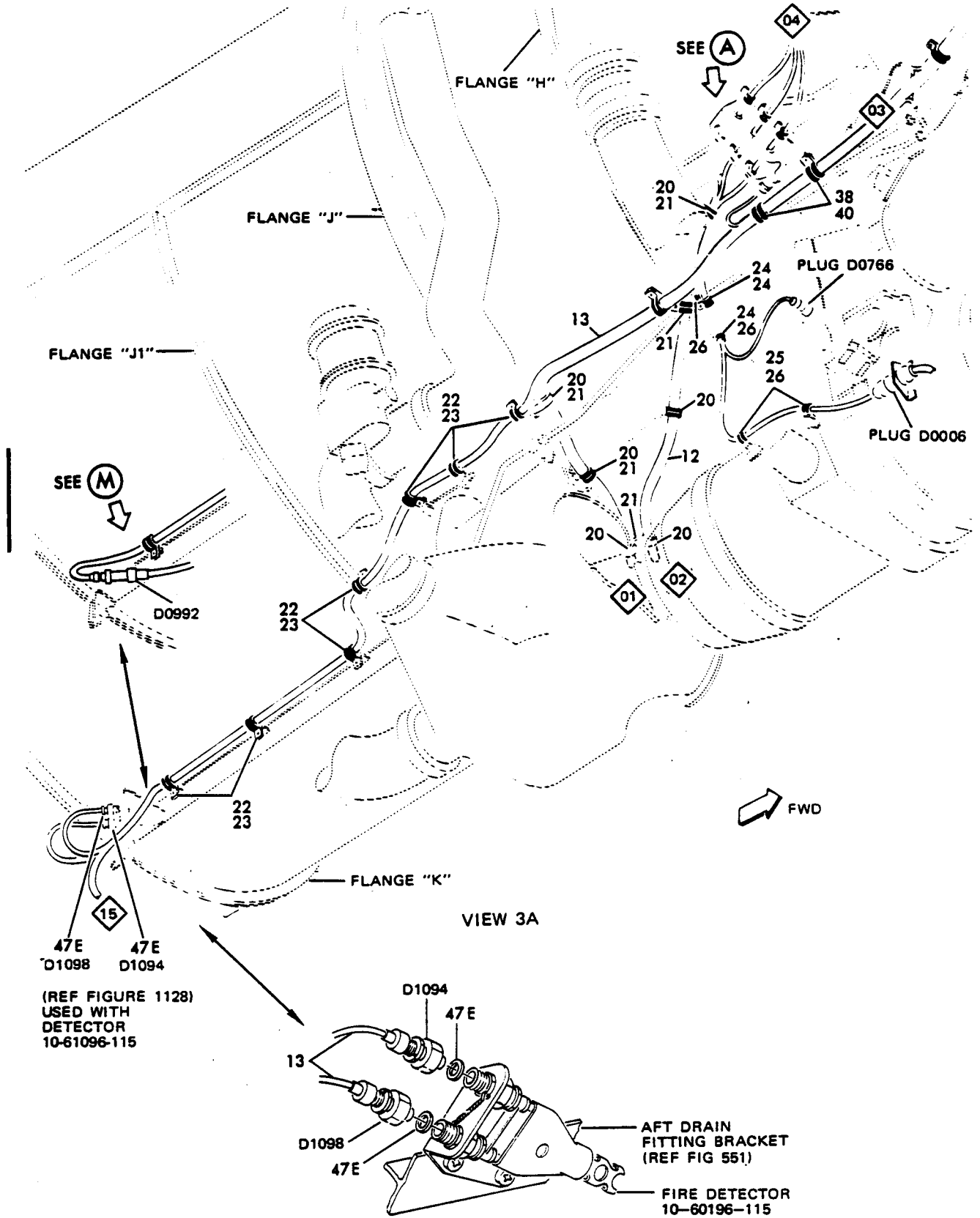
Jun 5/86



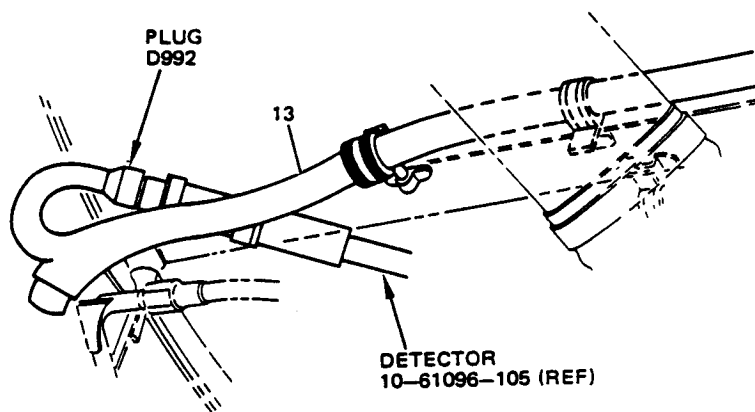
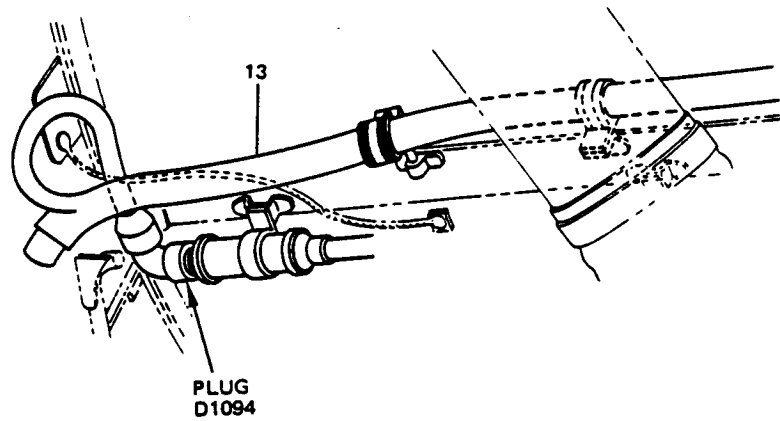
(L)

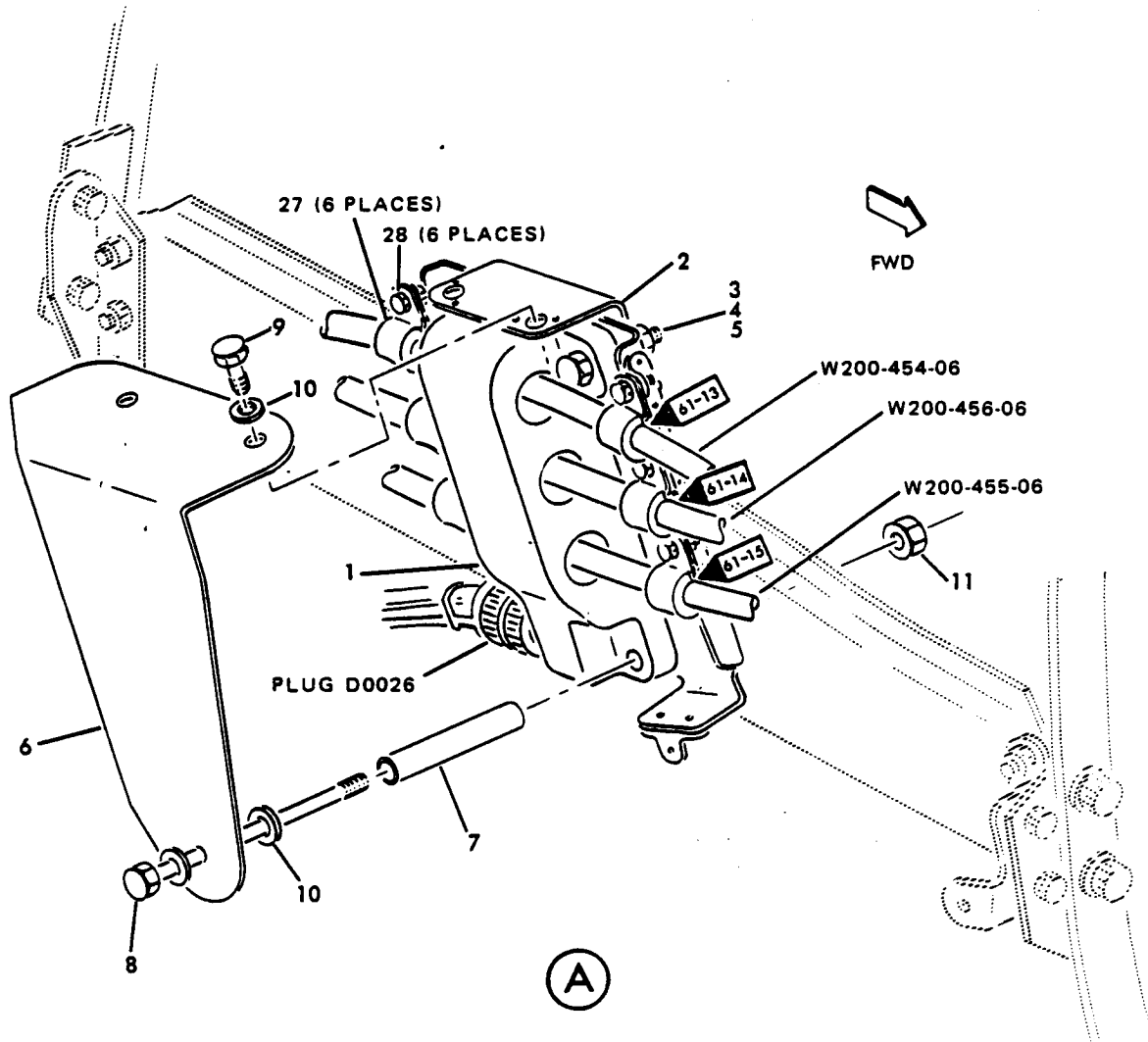


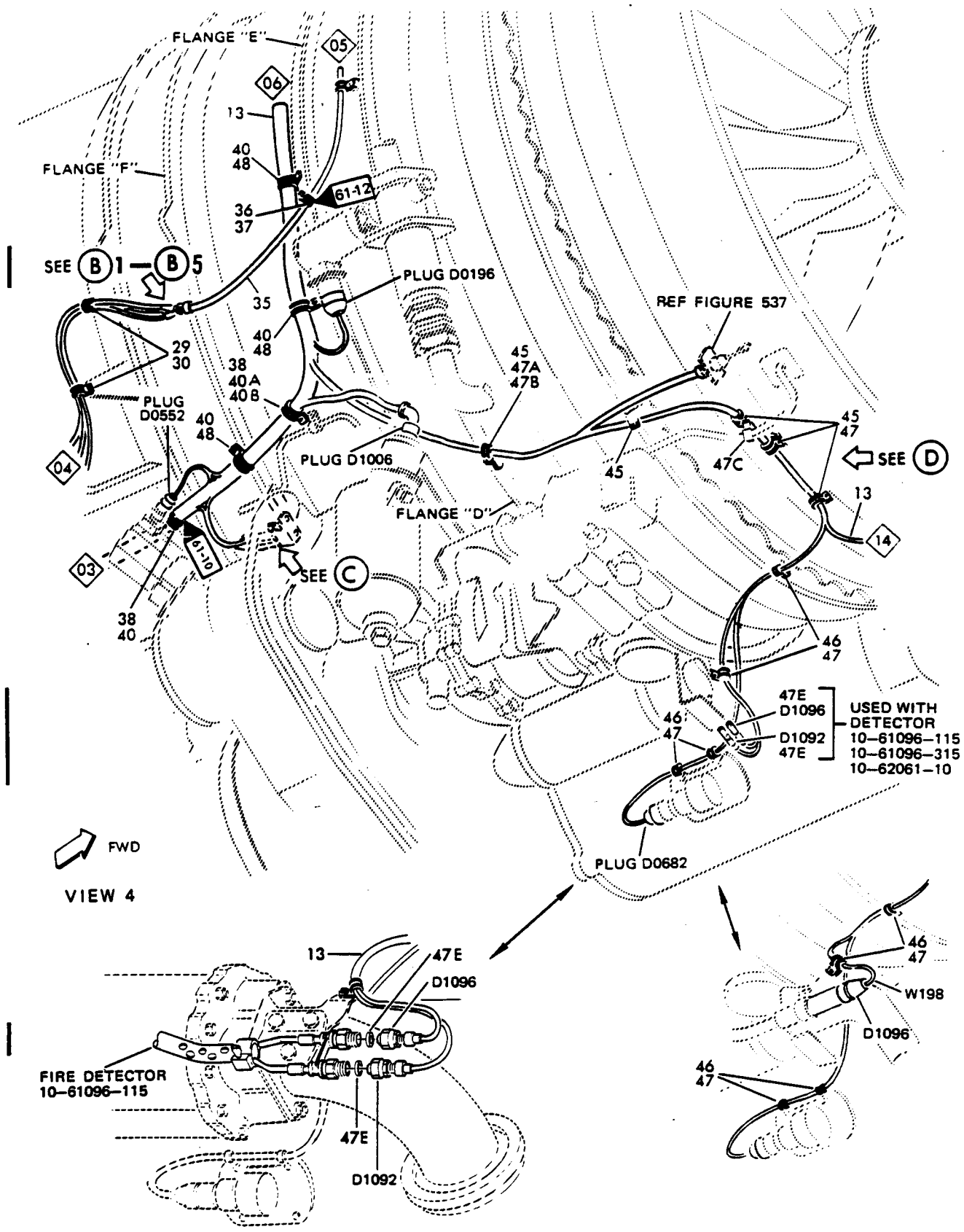
Electrical Installations
 Figure 525 (Sheet 7)



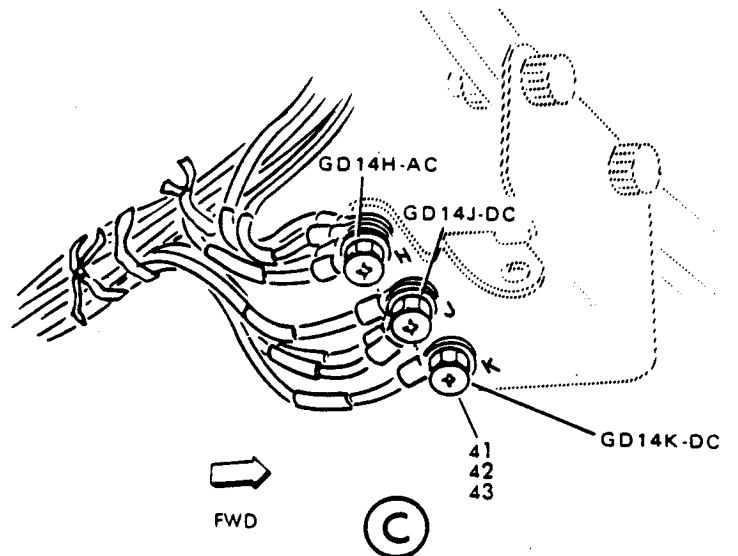
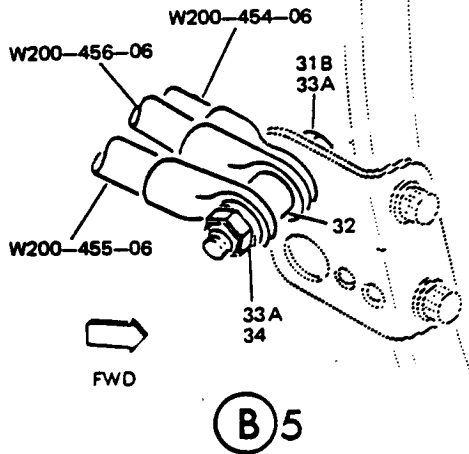
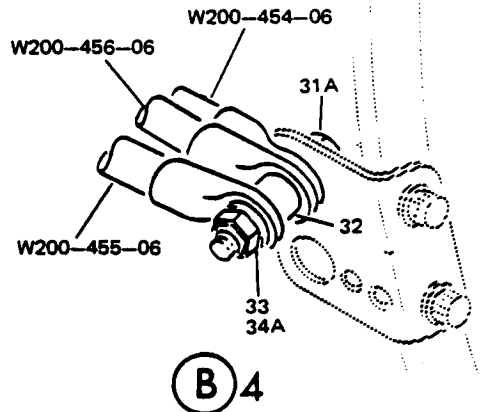
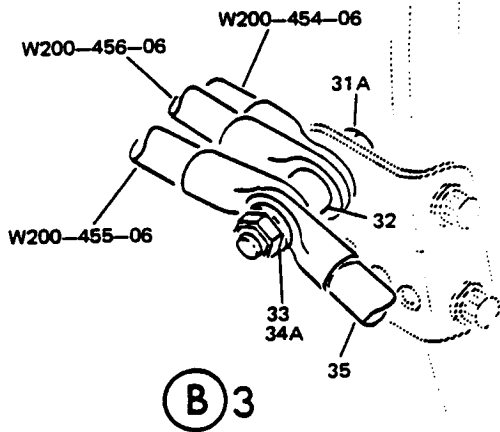
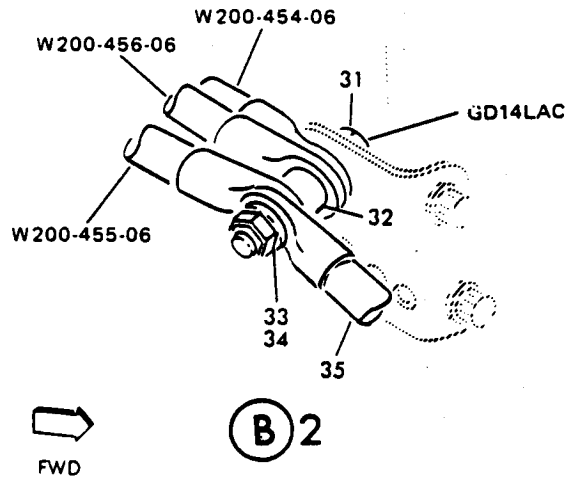
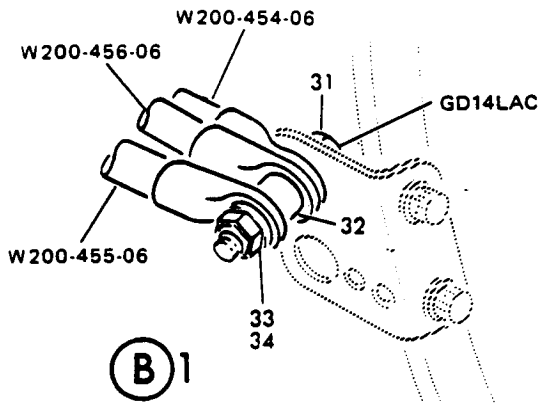
Electrical Installations
Figure 525 (Sheet 8)



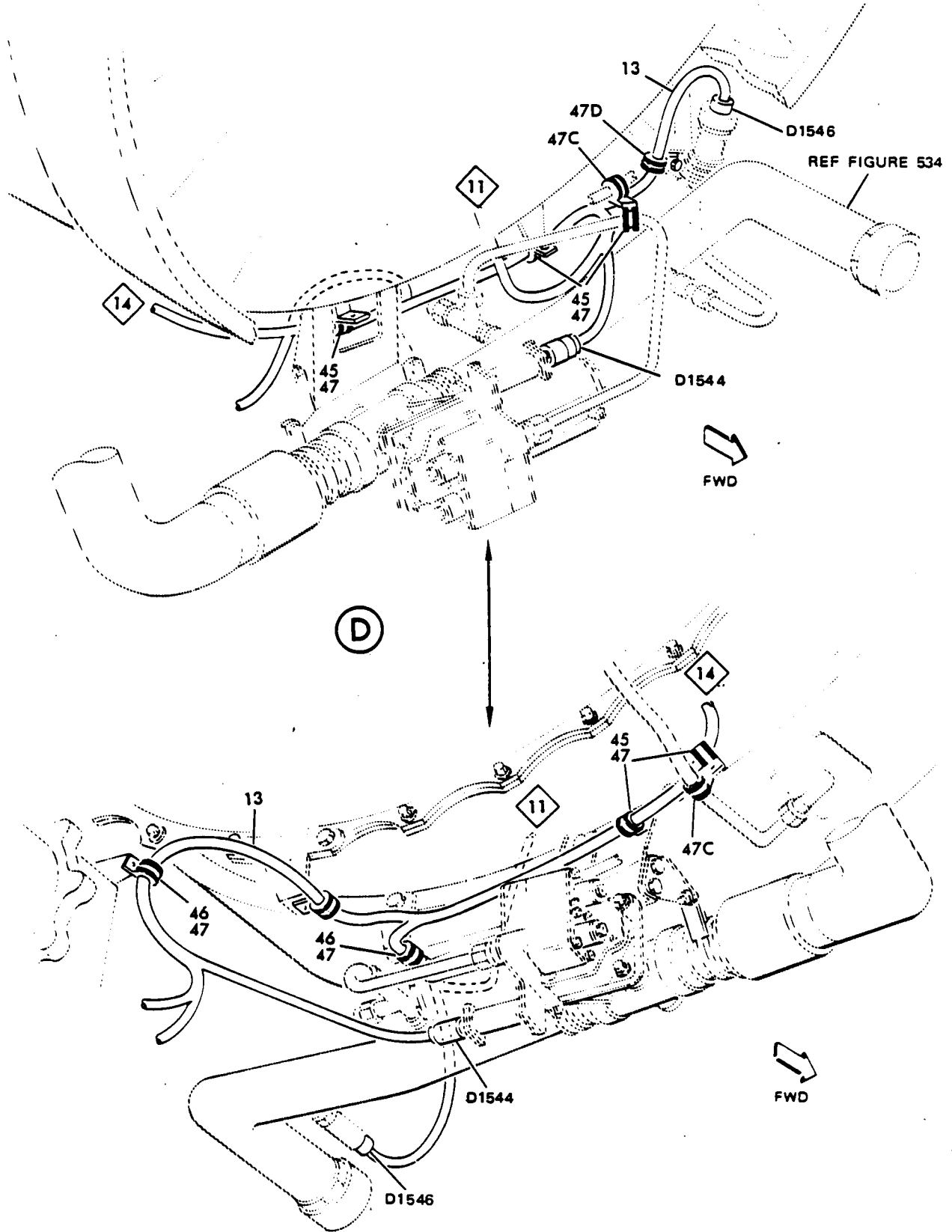


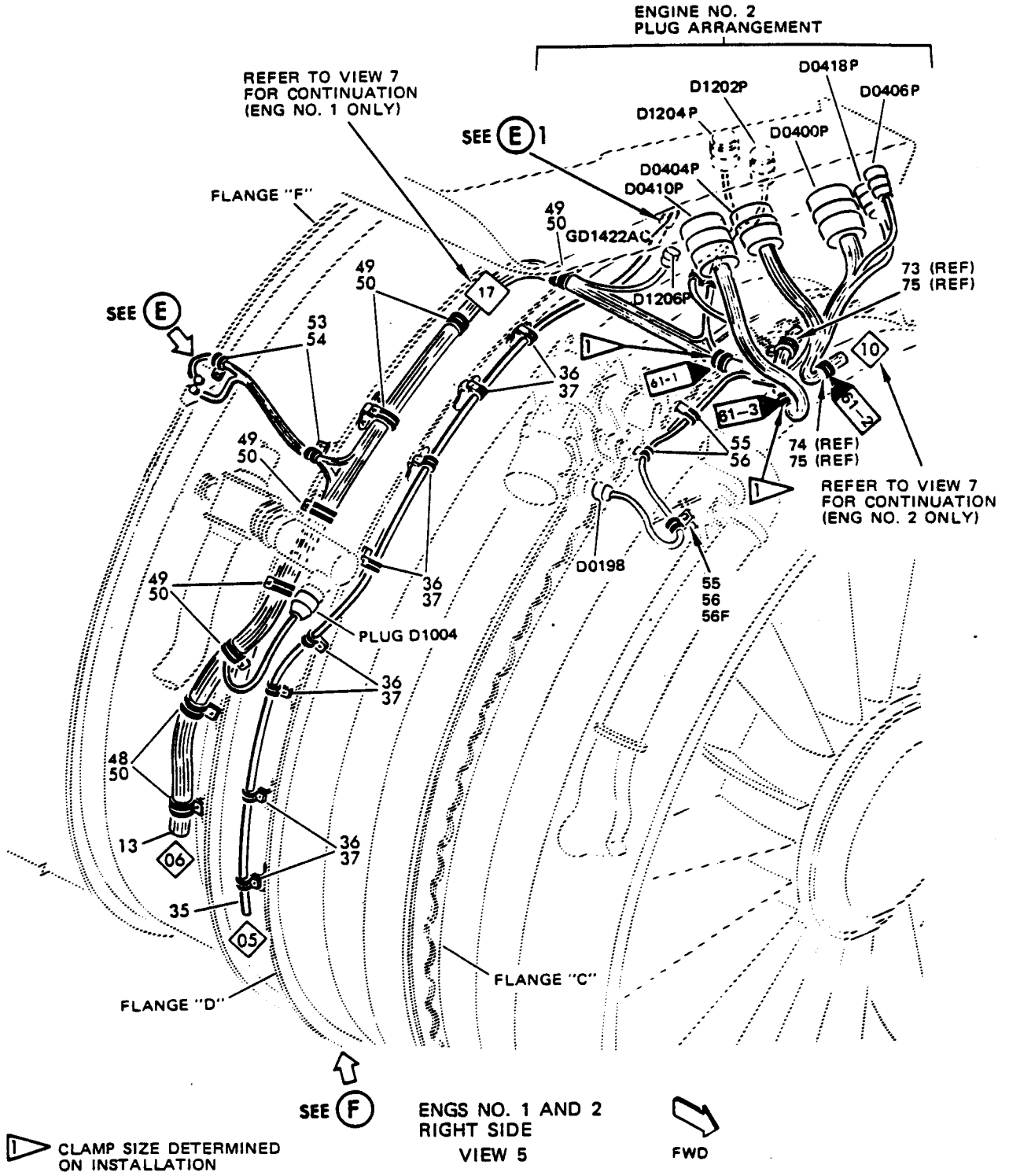


Electrical Installations
Figure 525 (Sheet 11)

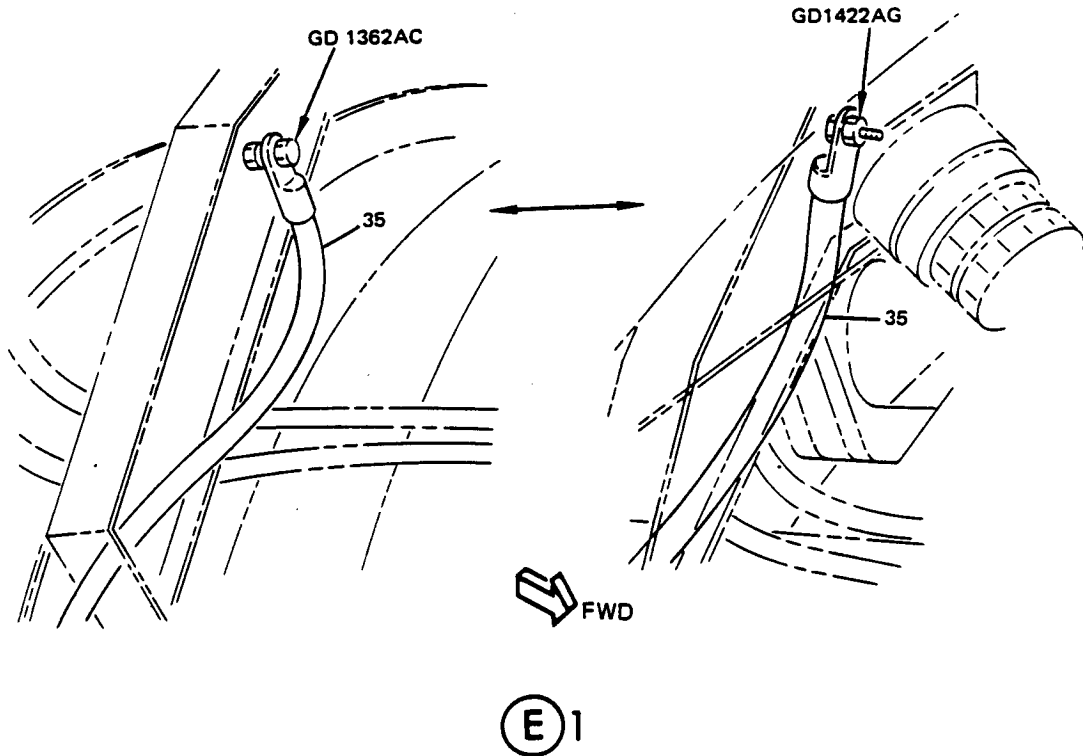
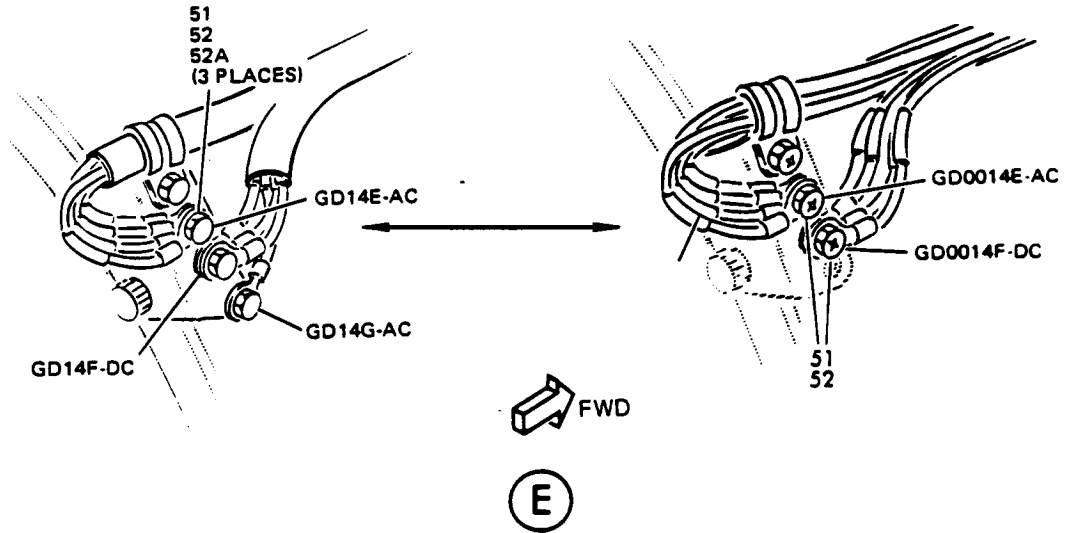


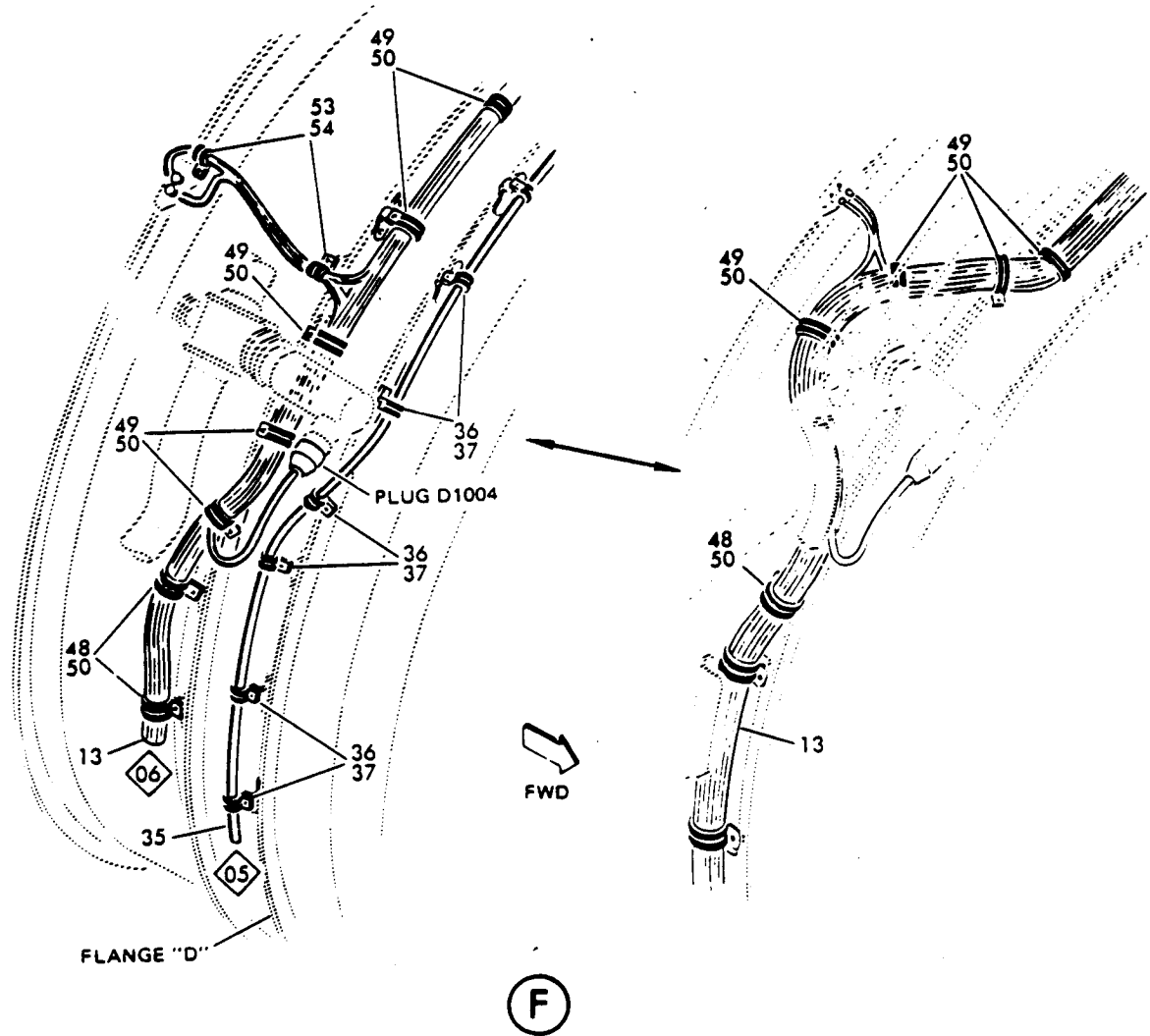
Electrical Installations
Figure 525 (Sheet 12)



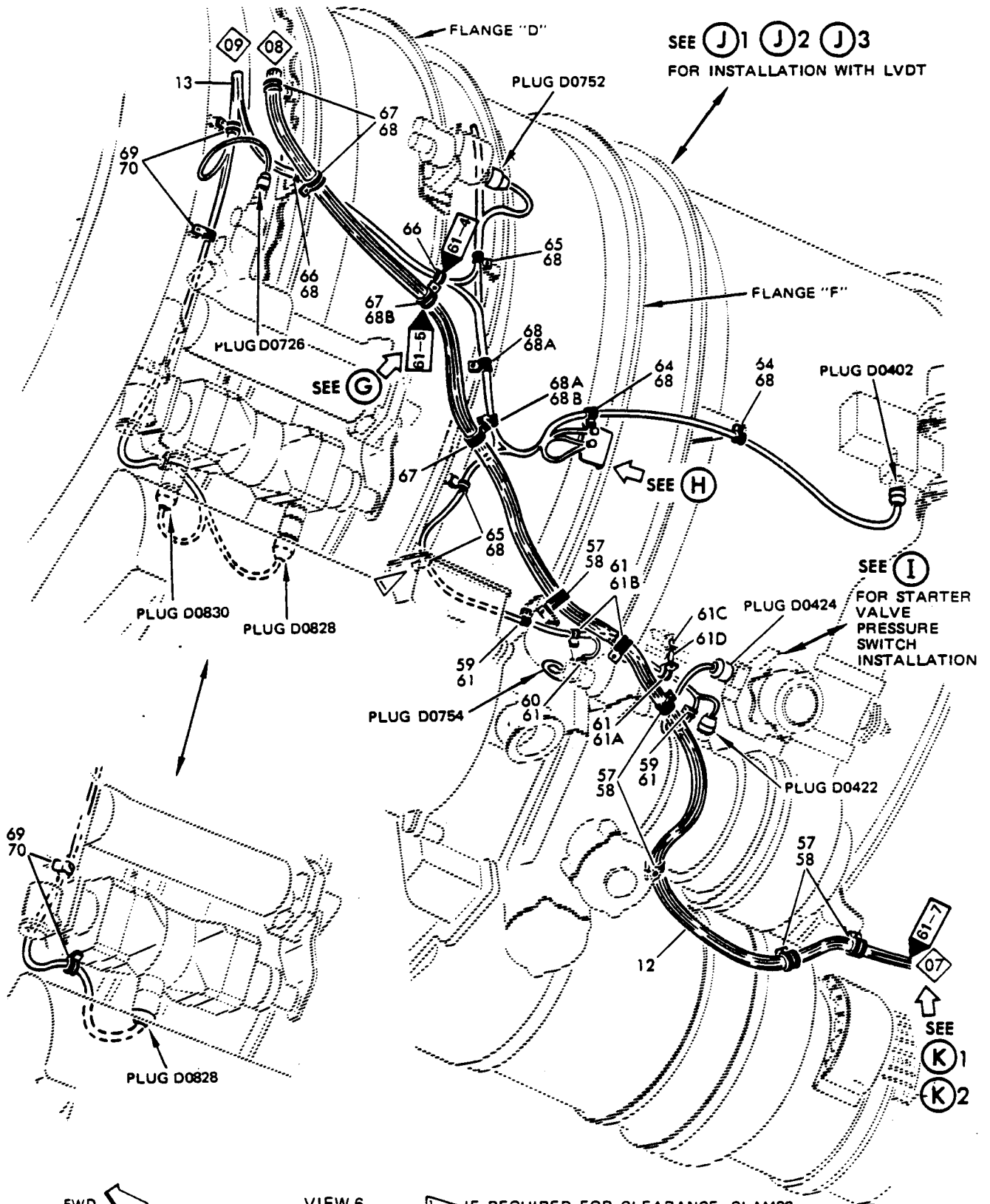


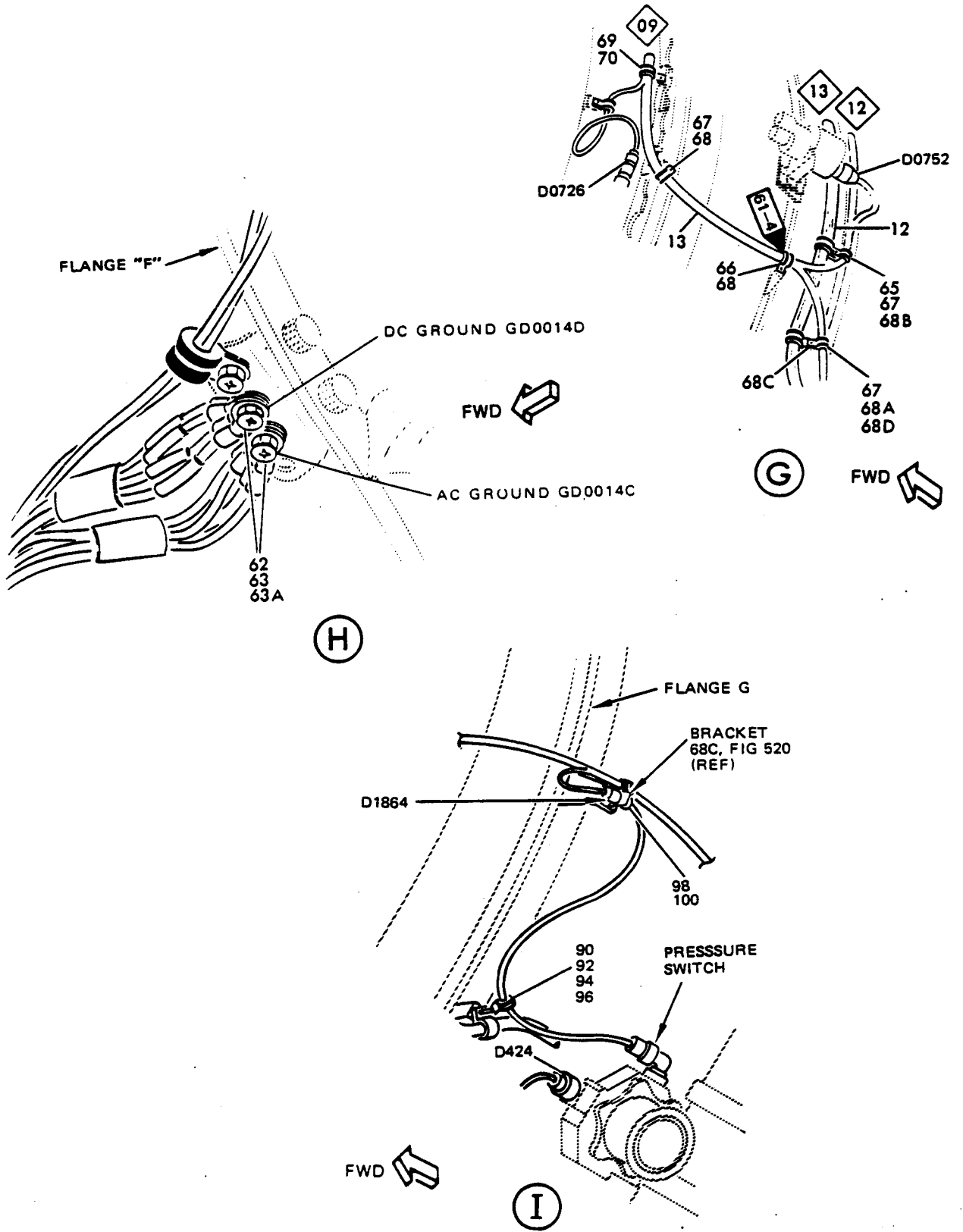
OVERHAUL MANUAL



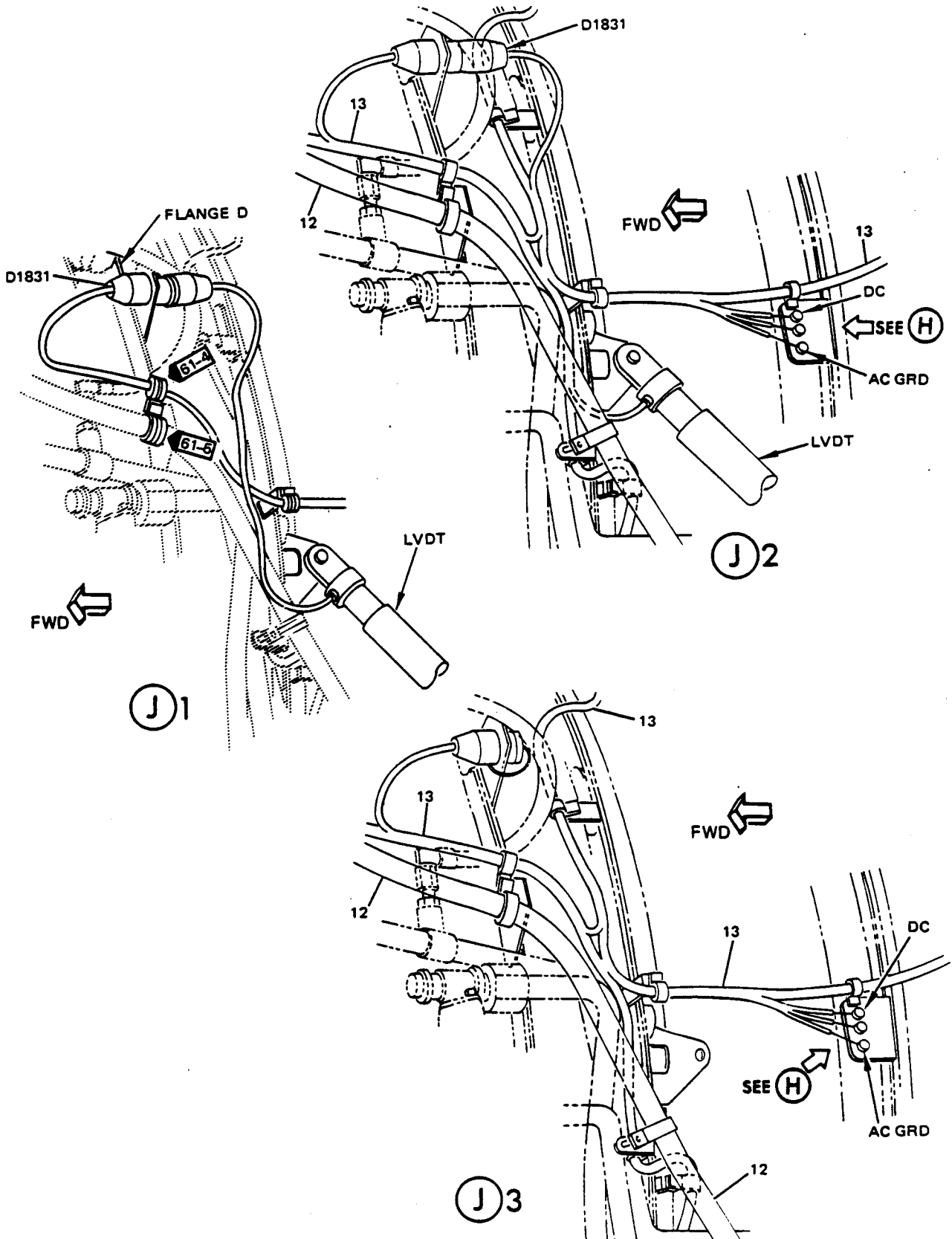


OVERHAUL MANUAL

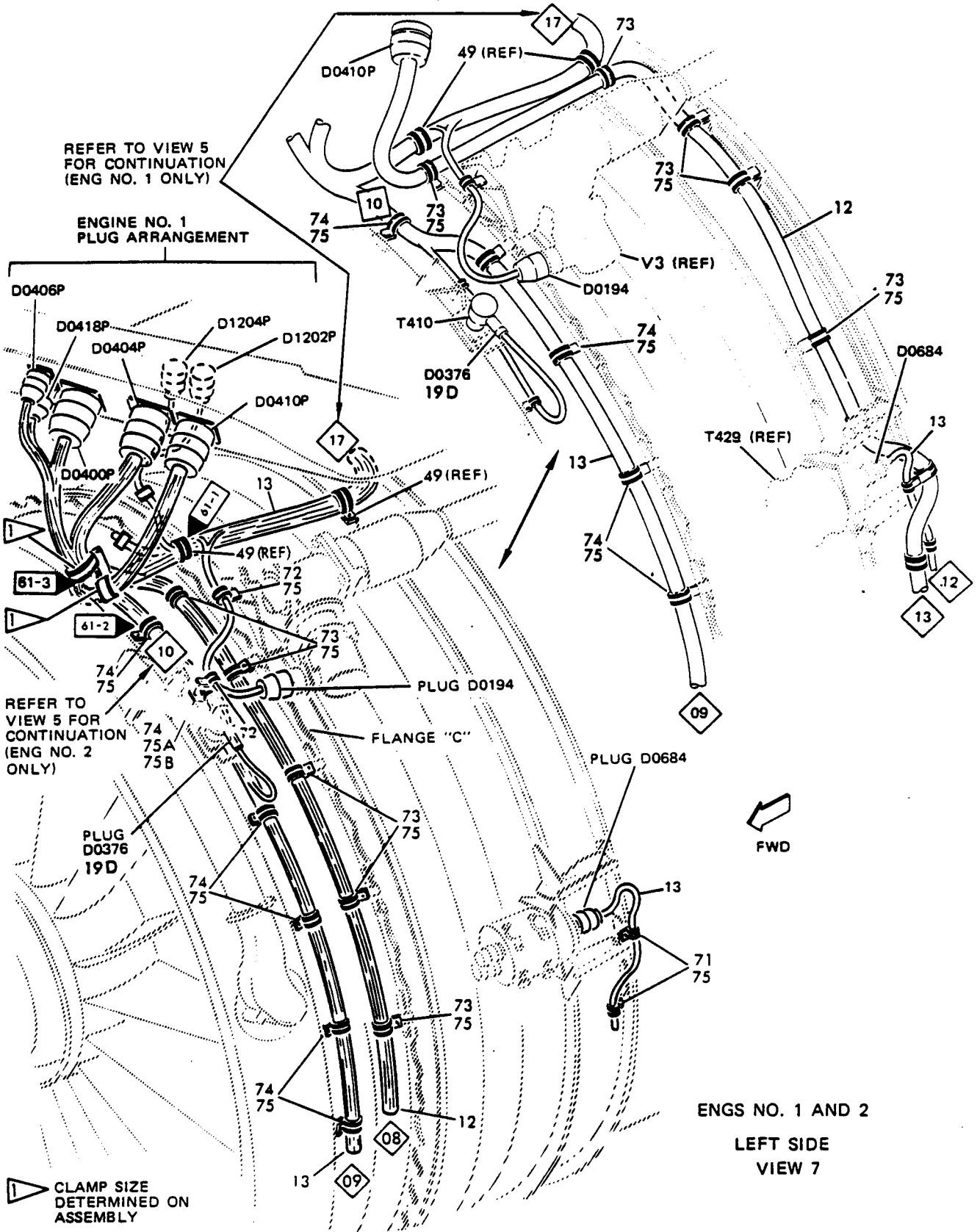




OVERHAUL MANUAL

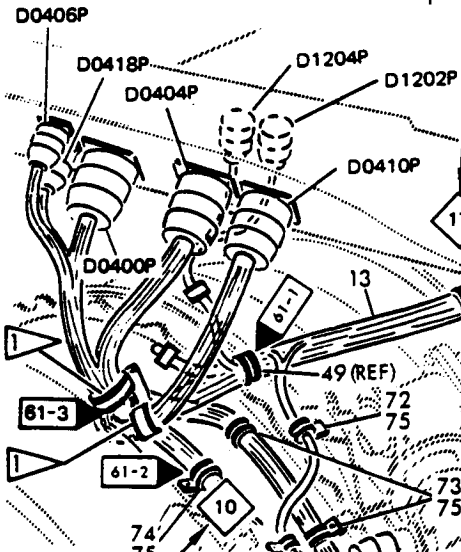


Electrical Installations
Figure 525 (Sheet 19)



REFER TO VIEW 5
FOR CONTINUATION
(ENG NO. 1 ONLY)

ENGINE NO. 1
PLUG ARRANGEMENT



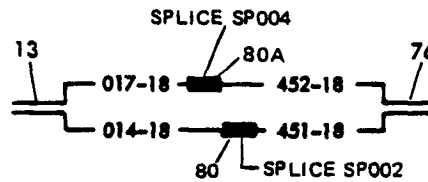
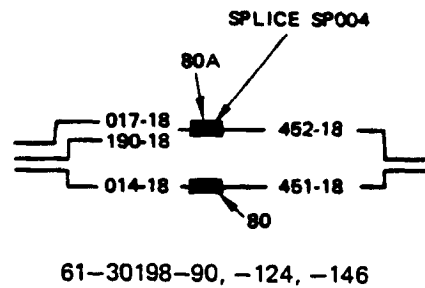
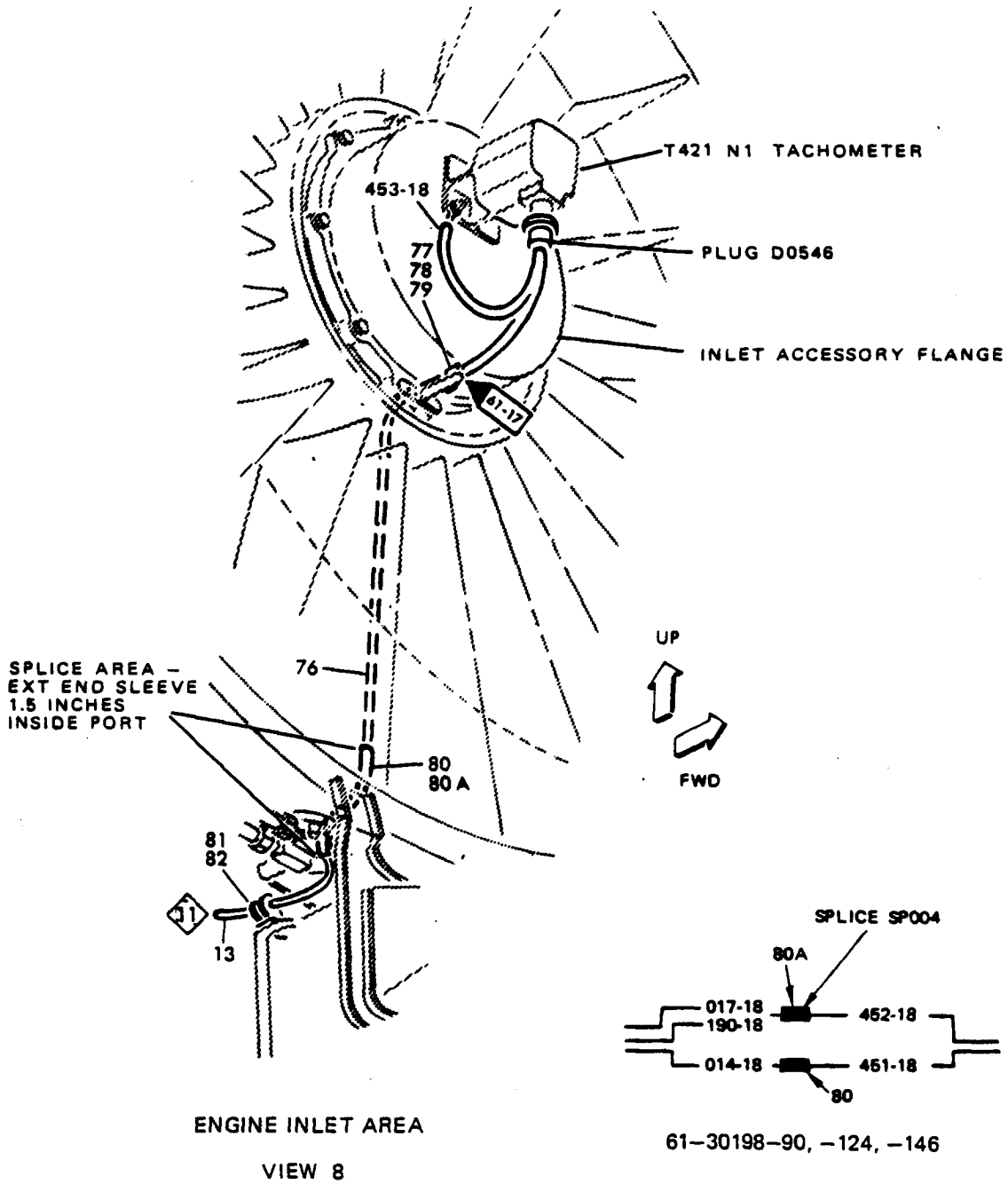
REFER TO
VIEW 5 FOR
CONTINUATION
(ENG NO. 2
ONLY)

PLUG D0376
19D

CLAMP SIZE
DETERMINED ON
ASSEMBLY

Electrical Installations
Figure 525 (Sheet 20)

OVERHAUL MANUAL



SPLICE SCHEMATIC

737

POWERPLANT BUILDUP



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
F. (1)	1	TRANSFORMER	10-61224-5		1
	2	BRACKET	69-52860-1		1
	3	SCREW	BACS12CB3-14		2
	4	WASHER	AN960D10L		2
	5	NUT	NAS679A3W		2
(2)	6	SHIELD, HEAT	65-65358-2		1
	7	SPACER	NAS43DD3-128		1
	8	SCREW	BACS12CB3-48		1
	9	SCREW	BACS12CB3-7		2
	10	WASHER	AN960D10L		4
	11	NUT	NAS679A3W		1
3)(a)	12	WIRE BUNDLE, AC GENERATOR	61-30200-001		1
	12	WIRE BUNDLE ASSY, AC GENERATOR	61-30200-003		1
	12	WIRE BUNDLE ASSY, AC GENERATOR	61-30200-005		1
	12	WIRE BUNDLE ASSY, AC GENERATOR	61-30200-007		1
	-12	WIRE BUNDLE, AC GENERATOR	61-30200-006		1
(b)	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-005		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-006		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-007		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-009		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0011		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0012		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0015		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0016		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0026		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-0027		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-028		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-029		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-030		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-031		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-032		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-033		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-034		1
	13	DELETED	61-30198-035		
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-036		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-037		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-038		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-039		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-040		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-044		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-045		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-047		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-054		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-056		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-057		1
	13	DELETED	61-30198-060		
	13	DELETED	61-30198-061		
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-062		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-064		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-065		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-066		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-067		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-068		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-069		1

737

POWERPLANT BUILDUP



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-070		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-071		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-072		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-073		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-074		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-075		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-076		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-077		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-079		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-080		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-081		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-082		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-083		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-084		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-085		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-088		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-089		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-090		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-091		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-092		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-093		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-097		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-099		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-100		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-101		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-105		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-107		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-110		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-111		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-112		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-113		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-114		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-115		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-116		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-117		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-118		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-119		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-120		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-121		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-122		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-123		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-124		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-125		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-126		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-127		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-128		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-129		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-130		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-131		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-132		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-133		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-134		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-135		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-136		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-137		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-138		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-139		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-140		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-141		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-142		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-143		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-144		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-145		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-146		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-147		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-148		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-149		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-150		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-151		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-152		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-153		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-154		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-155		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-156		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-157		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-158		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-159		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-160		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-161		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-162		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-163		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-164		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-165		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-166		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-167		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-168		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-169		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-170		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-171		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-172		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-173		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-174		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-175		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-176		1

737

POWERPLANT BUILDUP



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-177		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-178		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-179		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-181		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-185		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-191		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-193		1
	13	WIRE BUNDLE, ENGINE CONTROLS AND INSTR	61-30198-195		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-200		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-201		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-202		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-204		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-205		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-206		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-207		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-208		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-209		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-211		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-213		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-214		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-215		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-216		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-217		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-218		1

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-222		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-223		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-224		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-226		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-227		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-228		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-229		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-230		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-231		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-232		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-233		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-234		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-235		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-236		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-237		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-238		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-239		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-240		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-241		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-242		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-243		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-244		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-245		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-246		1

737

POWERPLANT BUILDUP



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-247		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-248		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-249		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-250		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-251		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-252		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-253		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-254		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-255		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-256		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-257		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-258		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-259		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-260		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-261		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-262		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-263		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-264		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-267		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-268		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-269		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-270		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-271		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-273		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-274		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-275		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-276		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-277		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-278		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-280		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-281		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-282		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-283		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-284		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-285		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-286		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-290		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-291		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-292		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-293		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-294		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-295		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-296		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-297		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-298		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-299		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-300		1

737

POWERPLANT BUILDUP



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-301		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-302		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-303		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-304		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-305		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-306		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-307		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-308		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-309		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-310		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-311		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-312		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-313		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-314		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-315		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-316		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-317		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-318		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-319		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-320		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-321		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-322		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-323		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-324		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-325		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-326		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-327		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-328		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-329		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-330		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-331		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-332		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-333		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-335		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-336		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-337		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-338		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-339		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-341		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-342		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-343		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-344		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-345		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-346		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-347		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-348		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-349		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-350		1

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-351		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-352		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-353		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-354		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-355		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-356		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-357		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-358		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-359		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-360		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-361		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-364		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-366		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-368		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-369		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-370		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-371		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-372		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-373		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-374		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-375		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-380		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-381		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-383		1

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-395		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-402		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-403		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-404		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-409		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-412		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-413		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-417		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-424		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-425		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-426		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-427		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-428		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-429		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-430		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-431		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-432		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-433		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-434		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-435		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-436		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-437		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-438		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-439		1

737

POWERPLANT BUILDUP

BOEING
COMMERCIAL JET
 OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-440		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-441		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-446		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-447		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-448		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-449		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-450		1
	13	WIRE BUNDLE, ENGINE CONTROL AND INSTR	61-30198-453		1
	13F	SPACER, BOTTOM	69-40075-2		1
	13H	SPACER, TOP	69-40075-4		1
	13J	SCREW	NAS514P-1032-16		2
	13L	WASHER	AN960-10L		2
	13N	NUT, SELF LOCKING	NAS679A3W		2
(c)	14	CLAMP	BACC10BH8RW		4
	15	SCREW	BACC12BH3-8		2
	16	CLAMP	BACC10BH7RW		AR
	17	CLAMP	BACC10BH5RW		5
	18	CLAMP	BACC10BH4RW		AR
	19	SCREW	BACS12CB3-7		AR
	19A	SCREW	BACS12CB3-5		4
	19B	CLAMP	BACC10BH2RW		1
	19C	SPACER	NAS43HT3-24		1
	19D	CONNECTOR CAP	101A052-3		2
(d)		NO PARTS REQUIRED			
(e)		NO PARTS REQUIRED			
(f)	20	CLAMP	BACC10BH8RW		7
	21	SCREW	BACS12CB3-7		6
(g)	22	CLAMP	BACC10BH7RW		AR
	22	CLAMP	BACC10BH3RW		AR
	23	SCREW	BACS12CB3-7		AR
(h)	24	CLAMP	BACC10BH7RW		2
	24A	SPACER	NAS43HT3-32		1
	25	CLAMP	BACC10BH6RW		2
	26	SCREW	BACS12CB3-(-)		4
(i)		NO PARTS REQUIRED			

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
(j)	27	CLAMP	BACC10BH4RW		6	
	28	SCREW	BACS12CB3-7		6	
(k)	29	CLAMP	BACC10BH8RW		2	
	30	SCREW	BACS12CB3-7		2	
	31	BOLT	BACB30LM6U9		1	
	31A	BOLT	AN6C-11		1	
	31B	BOLT	NAS566-39		1	
	32	SPACER	66-18526-1		1	
	33	WASHER	AN960C616		2	
	33A	WASHER	AN960-616L		1	
	34	NUT	BACN10JC6		1	
	34A	NUT	NAS679A6W		1	
	(k, 1)	35	WIRE, ENGINE GROUND	61-30202-001		1
		36	CLAMP	BACC10BH4RW		9
37		SCREW	BACS12CB3-7		9	
(l)	38	CLAMP	BACC10BH12RW		5	
	39	CLAMP	BACC10BH14RW		2	
	40	SCREW	BACS12CB3-7		6	
	40A	SPACER	NAS43HT3-56		1	
	40B	SCREW	BACS12CB3-32		1	
(m)	41	BOLT	BACS12CB3-8		3	
	42	WASHER	AN960D10L		3	
	43	NUT	NAS679A3W		3	
(n)	44	CLAMP	BACC10BH6RW		1	
	45	CLAMP	BACC10BH5RW		5	
	46	CLAMP	BACC10BH4RW		1	
	46A	SCREW	BACS12CB3-8		1	
	47	SCREW	BACS12CB3-7		6	
	47A	SPACER	NAS43HT3-56		1	
	47B	SCREW	BACS12CB3-32		1	
	47C	CLAMP	BACC10HS05		1	
	47D	CLAMP	BACC10DK2 (USED WITH VORTEX DISSIPATOR SYSTEM)		1	
(o)	47E	GASKET (KIDDE)	209592		4	
(p)	48	CLAMP	BACC10BH14RW		2	
	49	CLAMP	BACC10BH15RW		7	
	50	SCREW	BACS12CB3-7		9	
(q)		NO PARTS REQUIRED				
(r,s)	51	SCREW	BACS12CB3-8		2	
	51	SCREW	BACS12CB3-8 (ON ALTERNATE INSTL)		3	

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	52	NUT	NAS679A3W		2
	52	NUT	NAS679A3W (ON ALTERNATE INSTL)		3
	52A	WASHER	AN960D10L		2
	52A	WASHER	AN960D10L (ON ALTERNATE INSTL)		3
(t)	53	CLAMP	BACC10BH7RW		2
	54	SCREW	BACSL2CB3-7		2
(u)		NO PARTS REQUIRED			
(v)	55	CLAMP	BACC10BH5RW		3
	56	SCREW	BACSL2CB3-(-)		2
	56F	SPACER	NAS43HT3-56		1
	56G	CLAMP	BACC10HS04		1
(w)	57	CLAMP	BACC10BH8RW		6
	58	SCREW	BACSL2CB3-7		6
(x)	59	CLAMP	BACC10BH5RW		2
	60	CLAMP	BACC10BH2RW		2
	61	SCREW	BACSL2CB3-(-)		6
	61A	CLAMP	BACC10BH3RW		1
	61B	CLAMP	BACC10BH4RW		1
	61C	SPACER	NAS43HT3-16		1
	61D	SPACER	NAS43HT3-32		1
(y)	62	SCREW	BACSL2CB3-8		2
	63	NUT	NAS679A3W		2
	63A	WASHER	AN960D10L		2
(z)	64	CLAMP	BACC10BH3RW		2
	65	CLAMP	BACC10BH5RW		3
	66	CLAMP	BACC10BH7RW		2
	67	CLAMP	BACC10BH8RW		4
	68	SCREW	BACSL2CB3-7		9
	68A	CLAMP	BACC10BH6RW		2
	68B	SCREW	BACSL2CB3-8		2
	68C	SPACER	NAS43HT3-48		1
	68D	SCREW	BACSL2CB3-22		1
(aa)	69	CLAMP	BACC10BH6RW		2
	70	SCREW	BACSL2CB3-7		2
(ab)	71	CLAMP	BACC10BH4RW		1
	71A	CLAMP	BACC10BH3RW		
	72	CLAMP	BACC10BH5RW		2
	73	CLAMP	BACC10BH8RW		5

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	73	CLAMP	BACC10BH8RW		5
	74	CLAMP	BACC10BH9RW		6
	75	SCREW	BACS12CB3-7		13
	75A	SPACER	NAS43HT3-48		1
	75B	SCREW	BACS12CB3-22		1
(ad)	76	WIRE BUNDLE, N1 TACHOMETER GENERATOR	61-30196-001		1
	76	WIRE BUNDLE, N1 TACHOMETER GENERATOR	61-30196-003		1
	76	WIRE BUNDLE, N1 TACHOMETER GENERATOR	61-30196-005		1
	77	CLAMP	BACC10BH4RW		1
	78	SCREW	BACS12CB3-6		1
	79	NUT	NAS679A3W		1
(ae)	80	SPLICE	BACT12C20		2
(af)	81	CLAMP	BACC10BH3RW		1
	82	SCREW	BACS12CB3-7		1
(ag)	90	CLAMP	BACC10GE4		1
	92	WASHER	AN960D10L		3
	94	SPACER	NAS43HT3-80		1
	96	SCREW	NAS1801-3-30		1
	98	SCREW	NAS1801-04-5		4
	100	NUT	BACN10NW1		4

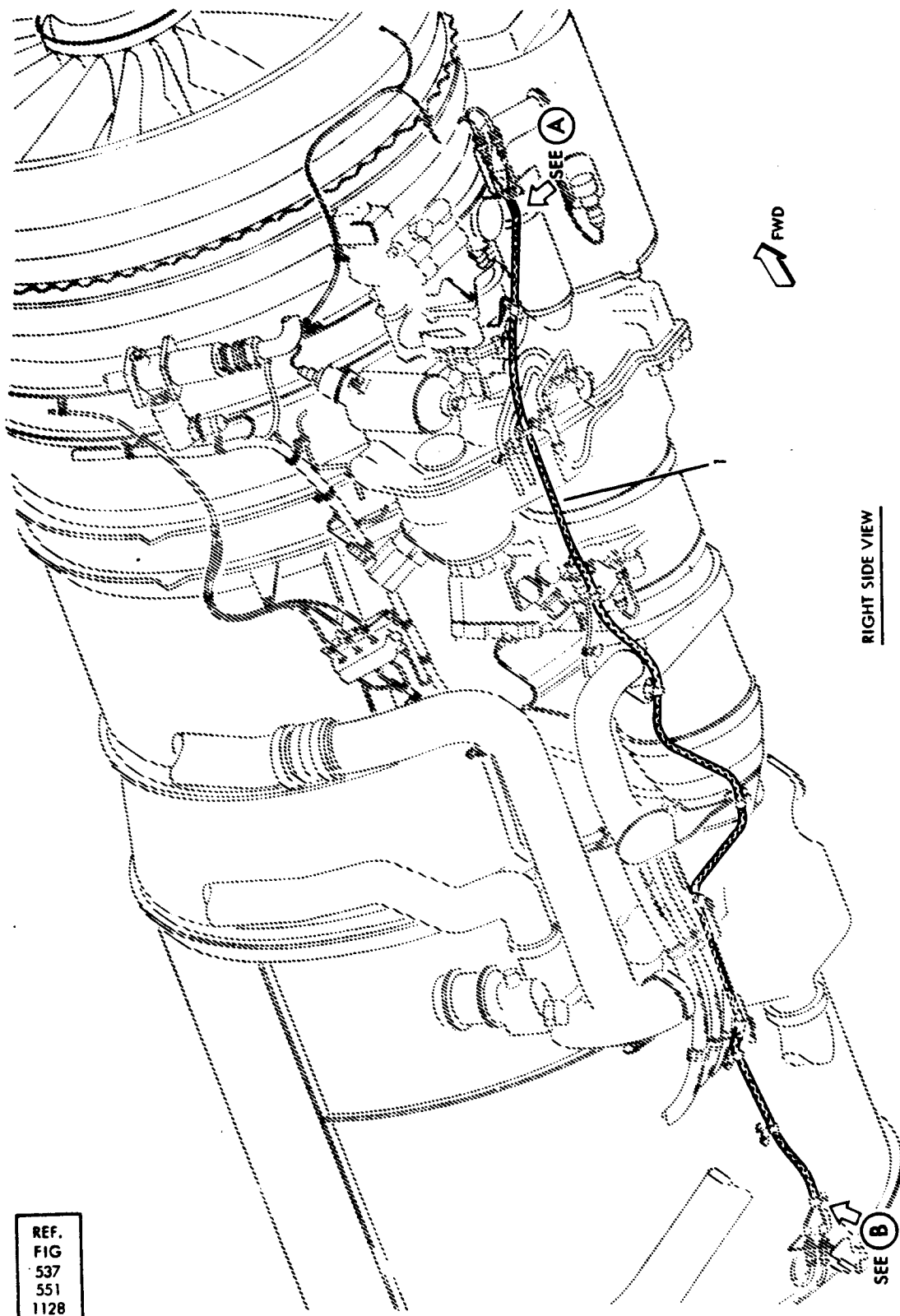
737
POWERPLANT BUILDUP

G. Install engine fire and overheat detector as follows: (Fig. 526)

- (1) Align sensing element (1) and clamp with five previously installed quick-release clamps.

NOTE: Sensing element must be placed in position at this time because of later inaccessibility. Instructions for completing installation are given with instructions for installing fuel heating air-exhaust duct (Fig. 537), and aft engine drain fitting assemblies (Fig. 551).

- (2) On Kidde system only, install jumper wires (2) and (3) using bolts (4), nuts (5) and washers (6). Tighten nut (5) to 65-70 lb-in. Make electrical bond check between jumper terminal and mounting bracket per 20-11-03, maximum allowable resistance is 0.001 ohms (1.0 milliohms).
- (3) Stow ends of sensing element to prevent damage.

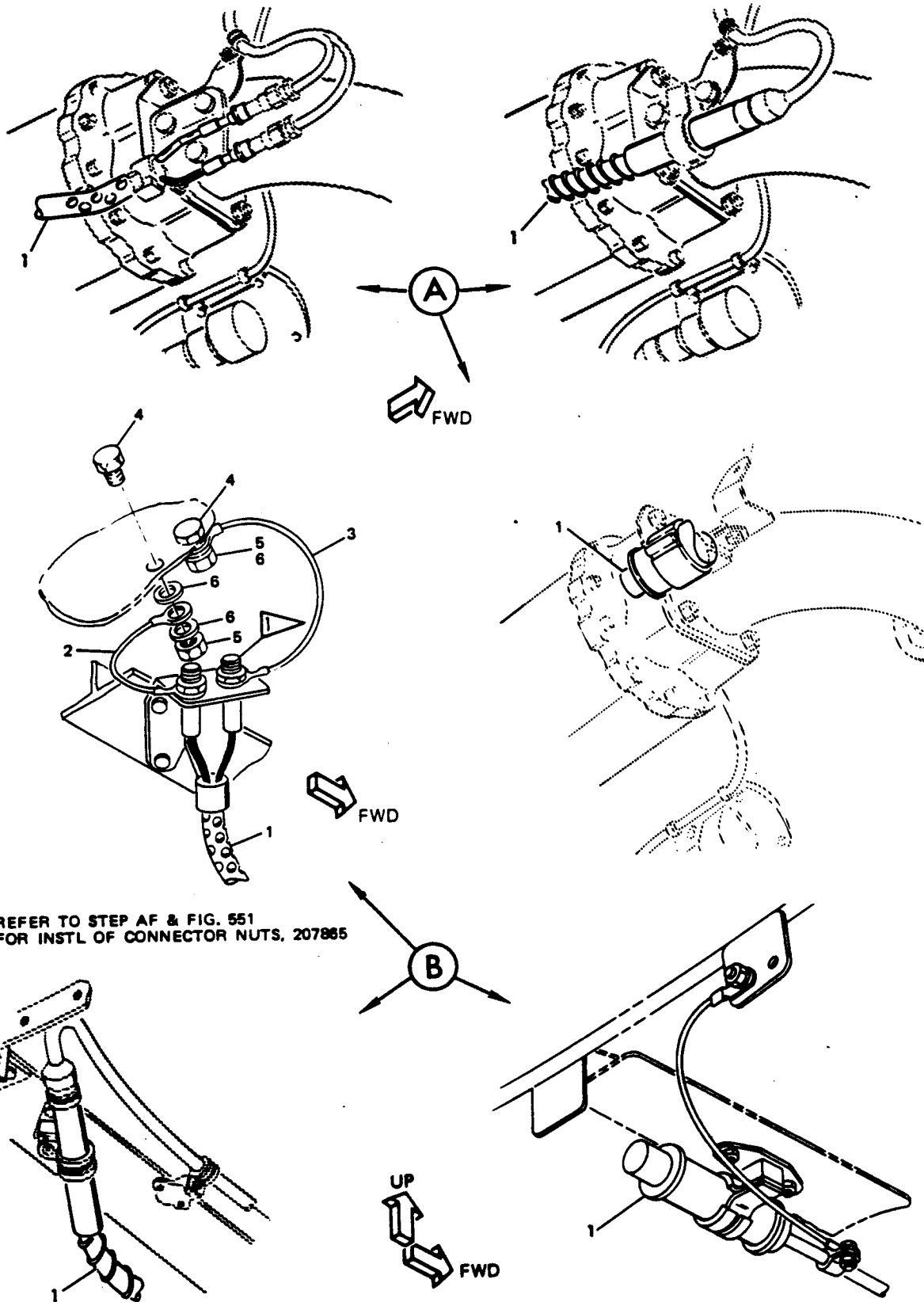


REF.
FIG
537
551
1128

Fire Detector Installation
Figure 526 (Sheet 1)

Jan 5/80

OVERHAUL MANUAL



△ REFER TO STEP AF & FIG. 551
FOR INSTL OF CONNECTOR NUTS, 207865

Fire Detector Installation
Figure 526 (Sheet 2)

737

POWERPLANT BUILDUP


BOEING
COMMERCIAL JET

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
G. (1)	1	ELEMENT, SENSING, FIRE DETECTOR (LINDBERG)	10-61096-105		1
	1	ELEMENT, SENSING, FIRE DETECTOR (KIDDE)	10-61096-115		1
	1	ELEMENT, SENSING, FIRE DETECTOR (KIDDE)	10-61096-315		1
	1	ELEMENT, SENSING, FIRE DETECTOR (KIDDE)	10-61096-615		1
	1	ELEMENT, SENSING, FIRE DETECTOR (SYSTRON-DONNER)	10-61096-35		1
	1	ELEMENT, SENSING, FIRE DETECTOR (SYSTRON-DONNER)	10-61096-40		1
	1	ELEMENT, SENSING, FIRE DETECTOR (KIDDE)	10-62061-10		1
(2)	2	JUMPER ASSY	BACJ40AD57-4		1
	2	JUMPER ASSY	BACJ40AD45-6		1
	3	JUMPER ASSY	BACJ40AD57-6		1
	4	BOLT	BACB30NE4-1		2
	5	NUT	BACN10JC4		2
	6	WASHER	AN960C416L		4
(3)		NO PARTS REQUIRED			

Fire Detector
Figure 526 (Sheet 3)

H. Install air-conditioning precooler assembly as follows: (Fig. 527)

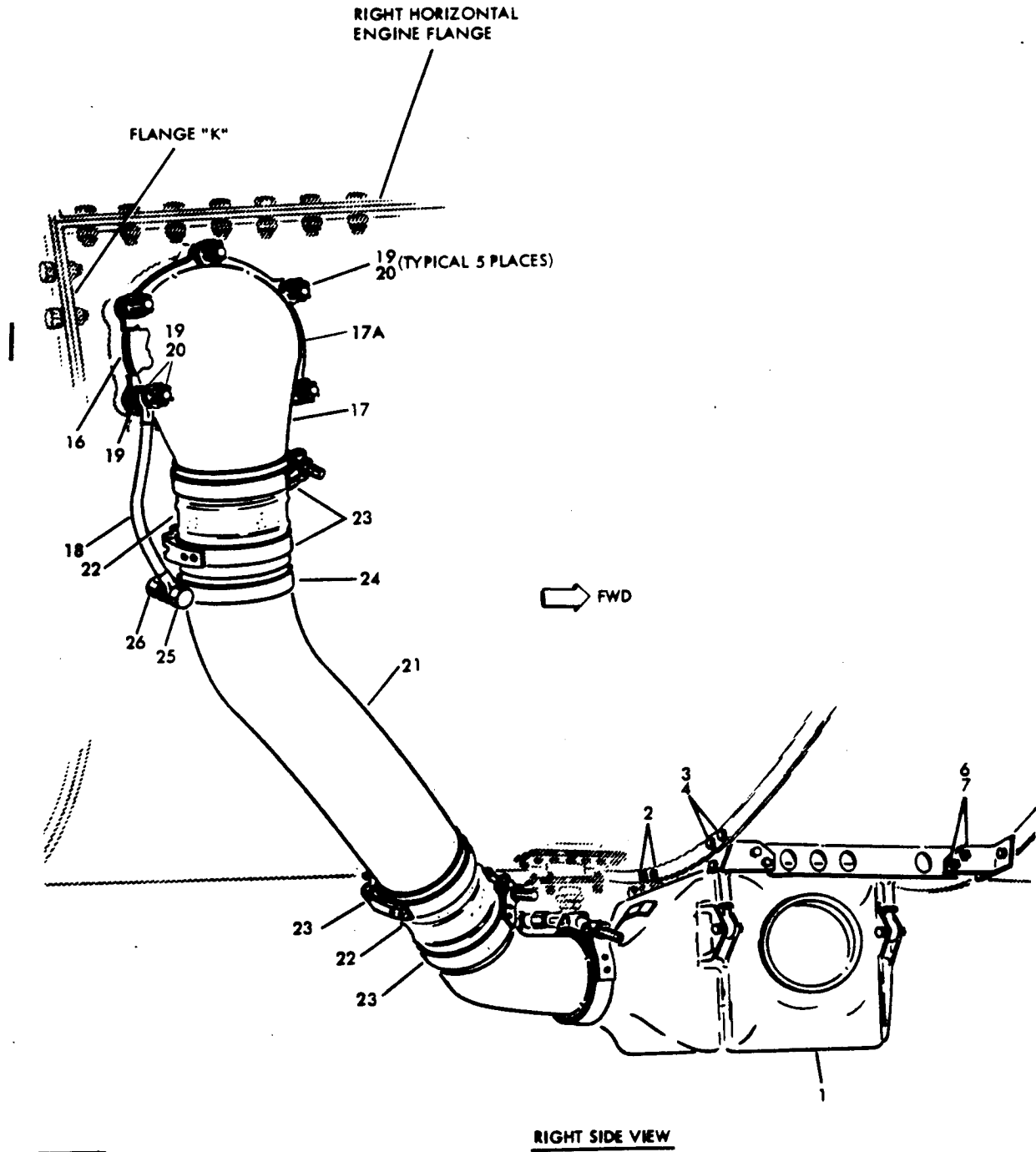
- (1) Locate precooler assembly (1) at bottom centerline of engine, between flanges J and J1. Attach brackets on aft end of precooler assembly to flange J1 with bolts (2 and 3) and nuts (4).

NOTE: Bolts (3) and nuts (4) must be installed and torqued to eliminate any gap between bracket assy and flange J1 before the two forward bolt holes in the bracket assy are drilled to match engine bracket hole locations.

- (2) Attach forward ends of brackets to existing brackets on flange J with bolts (5 and 6) and nuts (7).

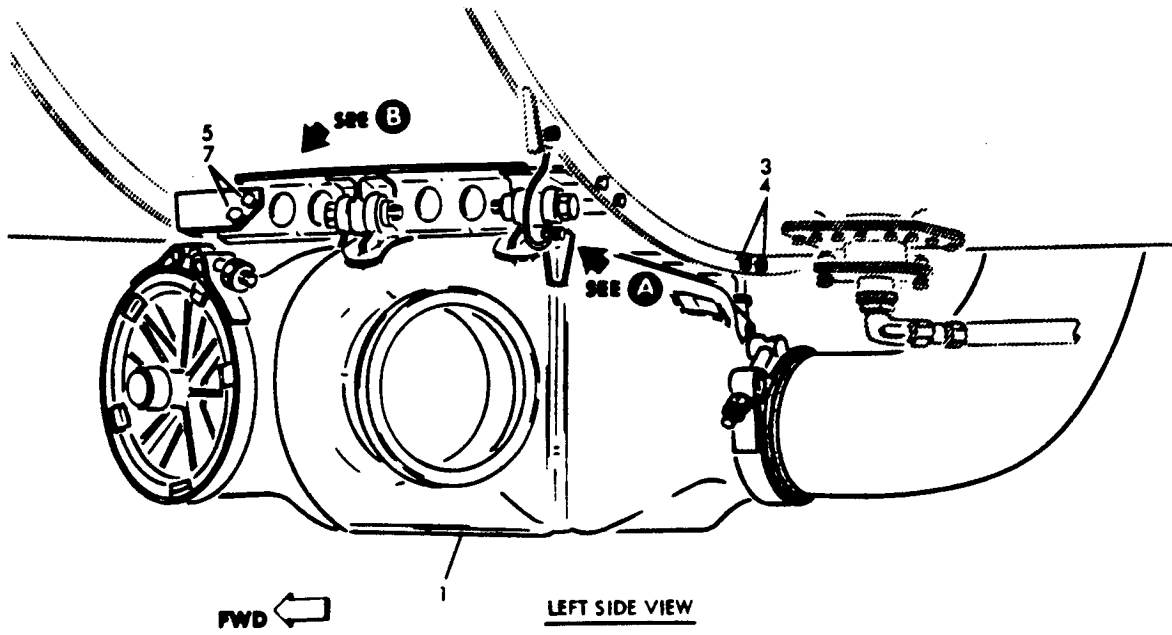
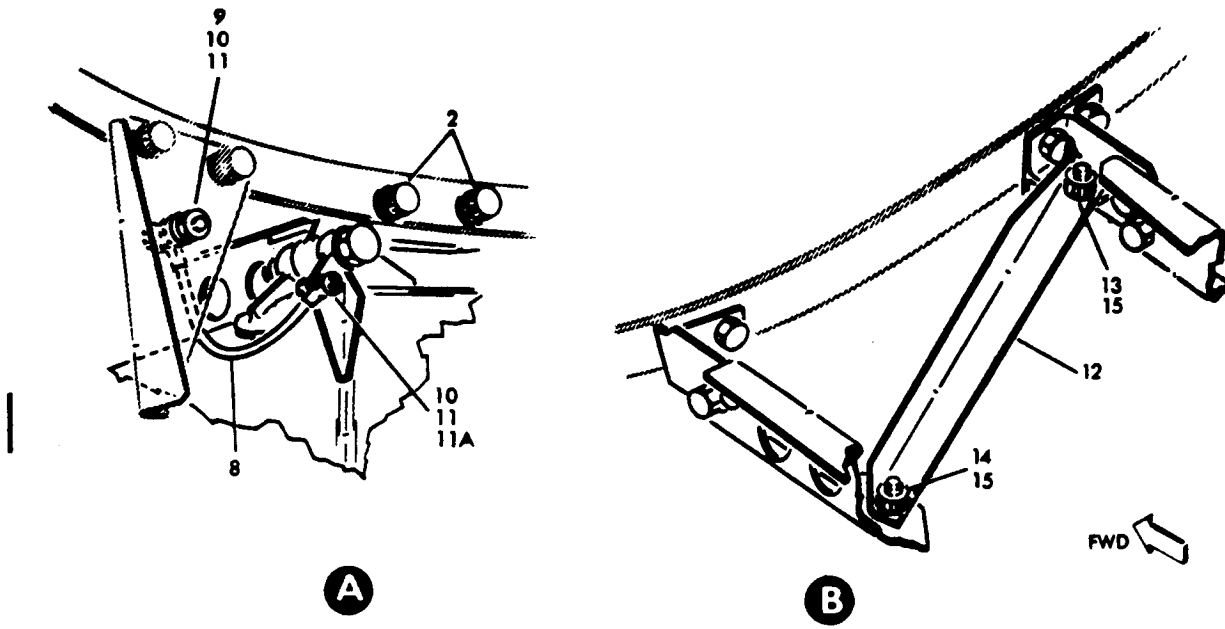
NOTE: The clearance between the engine case and the forward end of the Z shaped brackets should be within ± 0.030 inch of the clearance at the aft end of the brackets.

- (3) Lockwire bolts (2) together per 20-50-02.
- (4) Install bonding jumper (8) with screws (9), washers (10), and nuts (11). Tighten nuts (11) to 20-30 lb-in. Maximum allowable resistance is 0.001 ohms.
- (5) Install support angle (12) with bolts (13 and 14) and nuts (15).
- (6) Install P&W seal (16) on adapter (17), and install adapter on fan bleed port on right side of engine case aft of flange J1. When installing 65-46853-2 installation, install spacer (17A) between adapter (17) and engine port.
- (7) Locate bonding jumper (18) and install washers (19) and nut (20). Install washers (19) and nuts (20) five places.
- (8) Install duct (21) with two flexible ducts (22) and clamps (23). Tighten nut within torque range of 10 to 15 lb-in.
- (9) Install clamp (24) on duct (21), and attach free end of bonding jumper (18) between flanges of clamp with screw (25) and nut (26). Tighten nut (26) to 20-30 lb-in. Maximum allowable resistance is 0.001 ohms.



REF.
FIG
1113

Air-Conditioning Precooler Installation
Figure 527 (Sheet 1)



Air-Conditioning Pre-cooler Installation
Figure 527 (Sheet 2)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
H. (1)	1	PRECOOLER ASSEMBLY	(Preassembled in paragraph 3.I.)		Ref
	2	BOLT	BACB30CW4H2		2
	3	BOLT	BACB30CW4-4		2
	4	NUT	BACN10B-L4L		2
(2)	5	BOLT	NAS1304-2		2
	6	BOLT	NAS1304-3		2
	7	NUT	BACN10B-L4L		4
(3)		No parts required			
(4)	8	JUMPER, Bonding	MS25083-2BB4		1
	9	SCREW	NAS623-3-3		1
	10	WASHER	AN960-10L		6
	11	NUT	NAS679A3W		2
	11A S	CREW	NAS623-3-4		1
(5)	12	ANGLE, Support	69-42569-3		1
	13	BOLT	NAS1303-2		1
	14	BOLT	NAS1303-1		1
	15	NUT	NAS679C3W		2
(6)	16	SEAL	377129		1
	16	SEAL (OPT)	MS9386-161		1
	17	ADAPTER	65-50889-1		1
	17A S	PACER	69-62274-1		1
(7)	18	JUMPER, Bonding	MS25083-2BC8		1
	19	WASHER	AN960PD416		7
	20	NUT	BACN10B-L4L		6
(8)	21	DUCT	69-62261-1		1
	21	DUCT	65-53867-4		1
	22	DUCT, Flexible	BACD40D14-12-36		2
	23	CLAMP	BACCL0BN362L-R		4
(9)	24	CLAMP	AN735D56		1
	25	SCREW	BACSL2CB3-6		1
	26	NUT	NAS679A3W		1

I. Install 8th and 13th stage air conditioning manifolds as follows: (See figure 528.)

- (1) Install manifold assembly (1) and bracket assembly (2) on right-hand eighth stage bleed pad with bolts (3, 4, and 5) and washers (6 and 7). (See detail A.) Install with gasket supplied by engine manufacturer with bleed port cover.

NOTE: A maximum gap of 0.05 inches is allowed between manifold flange and bleed pad surfaces before installing bolts.

- (2) Lockwire all bolts.

- (3) Adjust thickness of shim (8) to close gap between bracket on manifold assembly (1) and existing bracket on flange "H". Coat shim with primer, Specification BMS 10-11, Type I, and install shim, bolts (9), and washers (10).

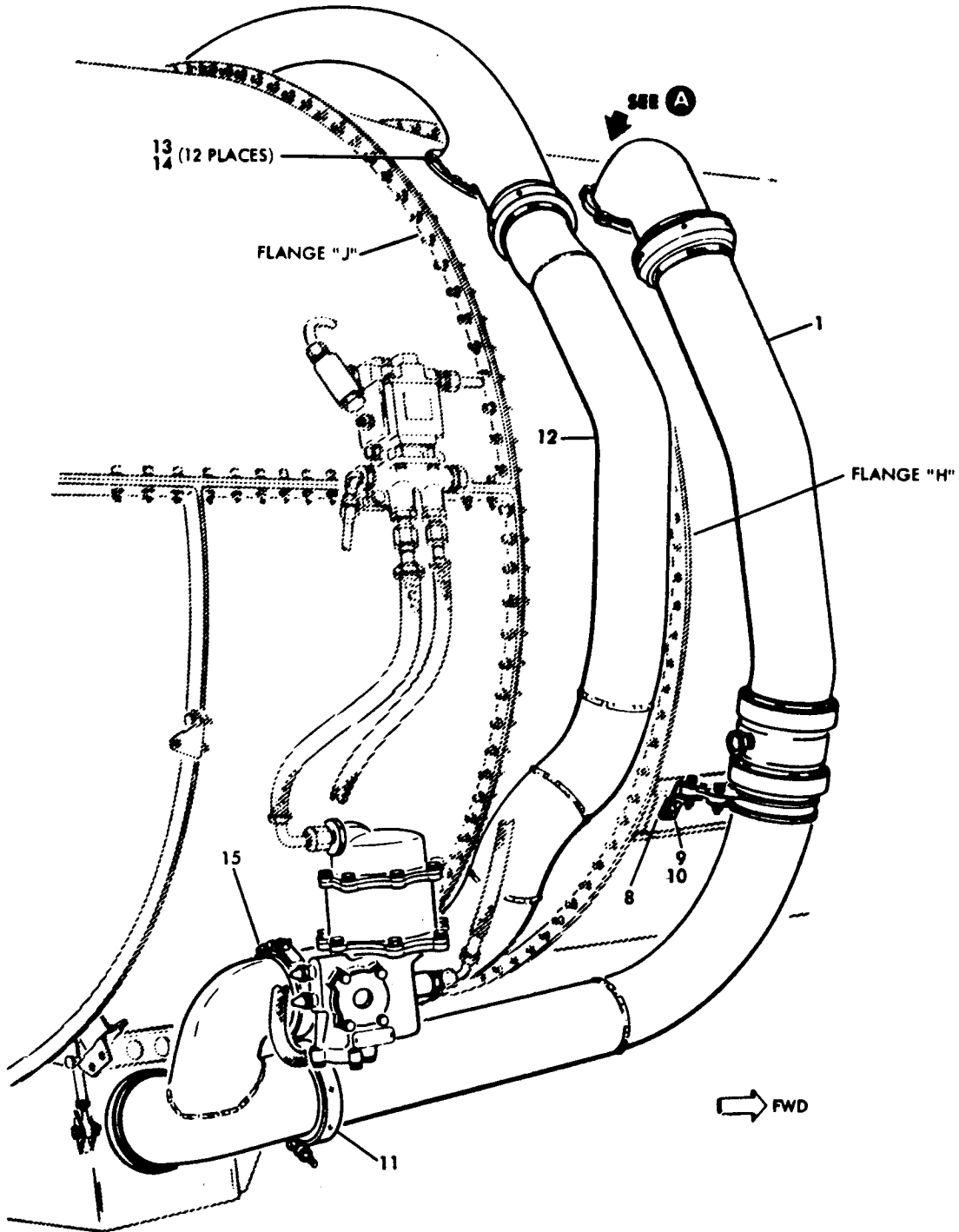
- (4) Attach manifold assembly (1) to air conditioning precooler with clamp (11).

- (5) Install manifold assembly (12) across top of engine with bolts (13) and washers (14). Install with gasket supplied by engine manufacturer with port covers.

NOTE: A maximum gap of 0.05 inches is allowed between manifold flange and bleed pad before installation of bolts.

- (6) Install lockwire MS20995N20 or MS20995NC20 in a continuous double-twist loop through all bolts.

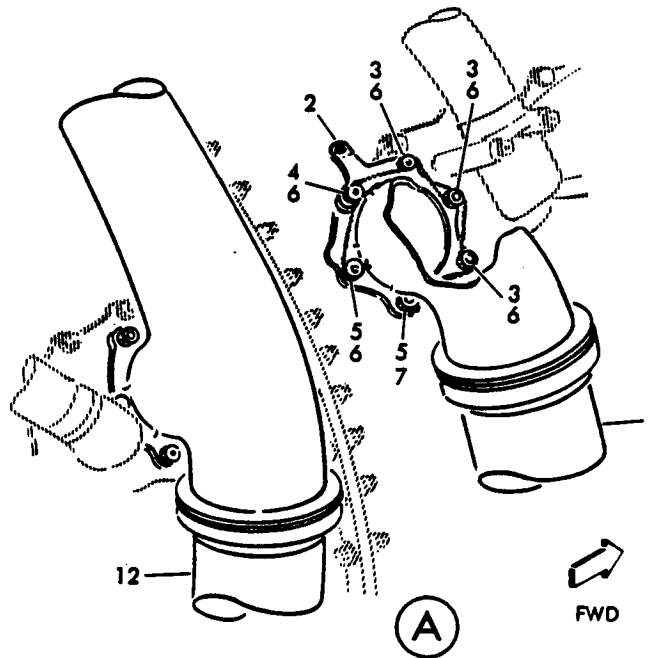
- (7) Attach manifold assembly (1) to shutoff valve on end of manifold assembly (12) with clamp (15).



RIGHT SIDE VIEW

REF.
FIG
506
507
1111

8th and 13th Stage Air-Conditioning Manifolds Installation
Figure 528 (Sheet 1)



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
I. (1)	1	MANIFOLD ASSY	(PREASSEMBLED IN PARAGRAPH 3.G.)		REF
	2	BRACKET ASSY	69-48235-4 *[1]		1
	3	BOLT	BACB30EJ4H2		3
	4	BOLT	BACB30EJ4H3		1
	5	BOLT	BACB30EJ4H7		2
	6	WASHER	BACW10BP4CT		5
	7	WASHER	BACW10BP4CT		1
(2)		NO PARTS REQUIRED			
(3)	8	SHIM	BACS40L12-28		1
	9	BOLT	NAS1104-6		2
	10	WASHER	AN960C416L		2
(4)	11	CLAMP	BACC10DU350Y		1
(5)	12	MANIFOLD ASSY	(PREASSEMBLED IN PARAGRAPH 3.G.)		REF
	13	BOLT	BACB30EJ4H3		12
	14	WASHER	BACW10BP4CT		12
(6)		NO PARTS REQUIRED			
(7)	15	CLAMP	BACC10DU300Y		1

*[1] Limited usage

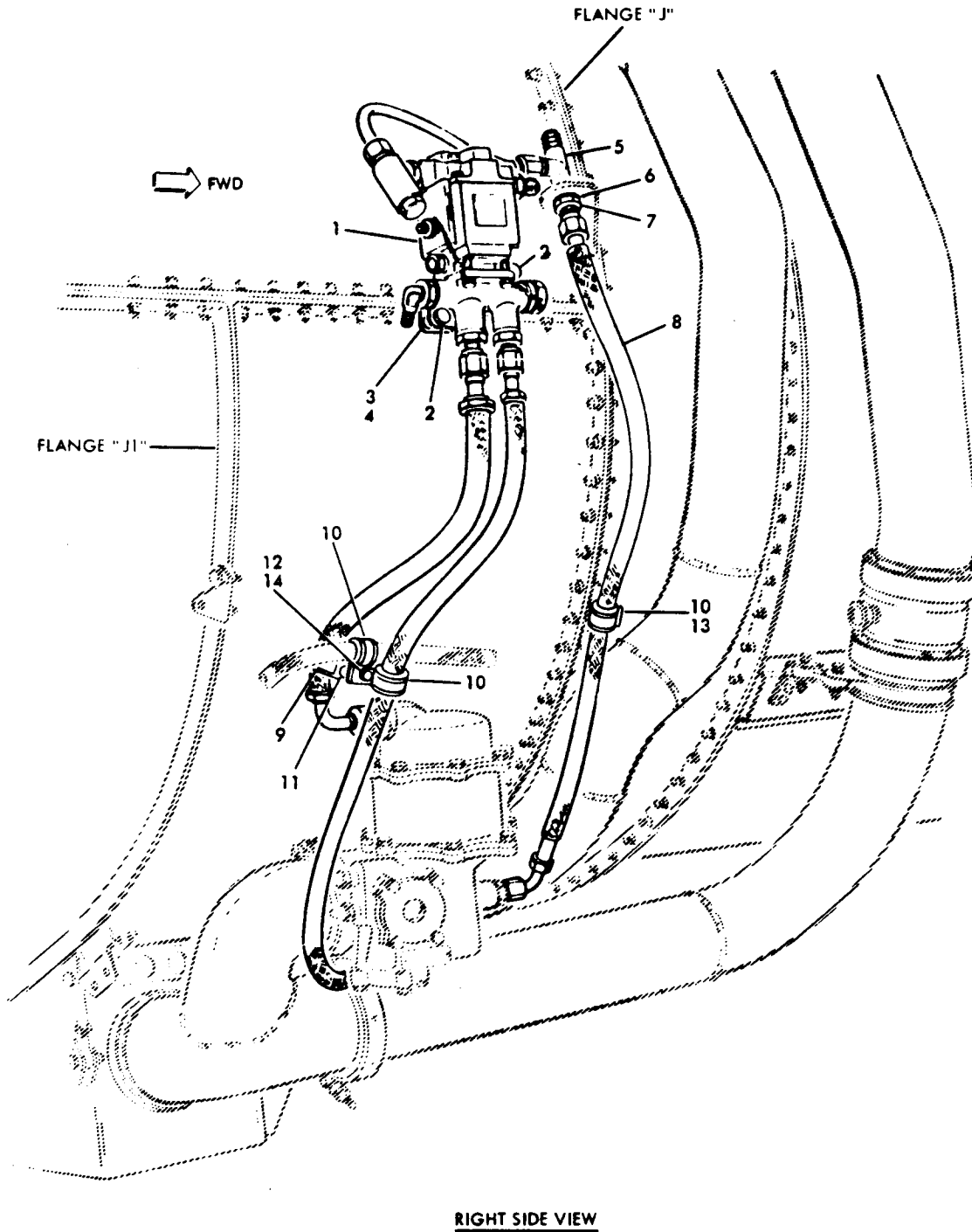
8th and 13th Stage Air Conditioning Manifolds Installation

Dec 5/86

Figure 528 (Sheet 2)

71-03-01
Page 556C

- J. Install pressure differential regulator and plumbing as follows: (See figure 529.)
- (1) Locate pressure differential regulator (1) on right side of engine, just aft of flange "J". Attach regulator to existing bracket with bolts (2), washers (3), and nuts (4).
 - (2) Connect forward hose on regulator to bottom fitting in valve in air-conditioning manifold. Tighten end fittings of hose.
 - (3) Connect aft hose on regulator to top aft fitting in valve in air-conditioning manifold. Tighten end fittings of hose.
 - (4) Install tee (5) on existing bracket on flange "J", just forward of regulator, with washer (6) and nut (7).
 - (5) Install hose (8) between bottom port of tee (5) and fitting in forward port of air-conditioning manifold shutoff valve. Tighten end fittings of hose.
 - (6) Attach tube assembly on regulator to aft port of tee (5). Tighten end fittings of tube assembly.
 - (7) Install clamps (9 and 10) with spacer (11), screws (12 and 13), and nut (14).
 - (8) Cap all open ports.



REF. FIG 513 1111

Pressure Differential Regulator Installation
Figure 529 (Sheet 1)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
J.					
(1)	1	REGULATOR, PRESSURE DIFFERENTIAL	(PREASSEMBLED IN PAR. 3.L.)		REF
	2	BOLT	NAS1104-15		2
	3	WASHER	AN960-416L		2
	4	NUT	BACN10JC4		2
(2)		NO PARTS REQUIRED			
(3)		NO PARTS REQUIRED			
(4)	5	TEE	MS21912-6		1
	6	WASHER	AN960-916		1
	7	NUT	AN924-6		1
(5)	8	HOSE	16228 (AVICA)		1
(6)		NO PARTS REQUIRED			
(7)	9	CLAMP	BACC10HS12		1
	10	CLAMP	BACC10HS08		3
	11	SPACER	NAS43HT3-32		1
	12	SCREW	BAC12CB3-12		1
	13	SCREW	BAC12CB3-8		1
	14	NUT	BACN10JC4		1
(8)		NO PARTS REQUIRED			

K. Install hydraulic system components as follows (Fig. 530):

(1) Fill hydraulic pump assembly (1) with hydraulic fluid BMS 3-11. Fill pump through MS21902-6 union in top of pump. On Vickers pumps only, lubricate shaft splines and cavity, using grease MIL-G-21164A.

(2) Place gasket (2) over studs on hydraulic pump mounting pad.

(3) Attach pump assembly (1) to engine as follows:

NOTE: If pump P/N 10-60470-7 or 10-61794-1 is used, perform steps (a), (b) and (c) only. If any other pump is used, perform step (d) only.

(a) Lubricate O-rings with BMS 3-11, Type 1.

(b) Install packing (5) and adapter (6), and attach with washers (4) and nuts (3).

(c) Attach pump assembly to adapter with coupling (7). Tighten coupling P/N 55759 to 115-125 pound-inches. Tighten clamp P/N 22807 to 50-60 pound-inches. Tighten mounting clamp P/N 374105 to 45-55 pound-inches.

(d) Locate pump assembly and attach with washers (4) and nuts (3).

(4) Install coupling halves (8 and 9) on existing bracket assembly just aft of flange J on horizontal centerline of engine. Attach couplings with nut (10).

(5) Install coupling halves (11 and 12) on bracket with nut (13).

(6) Install coupling halves (14 and 15) on bracket with nut (16).

(7) Lockwire couplings together per 20-50-02.

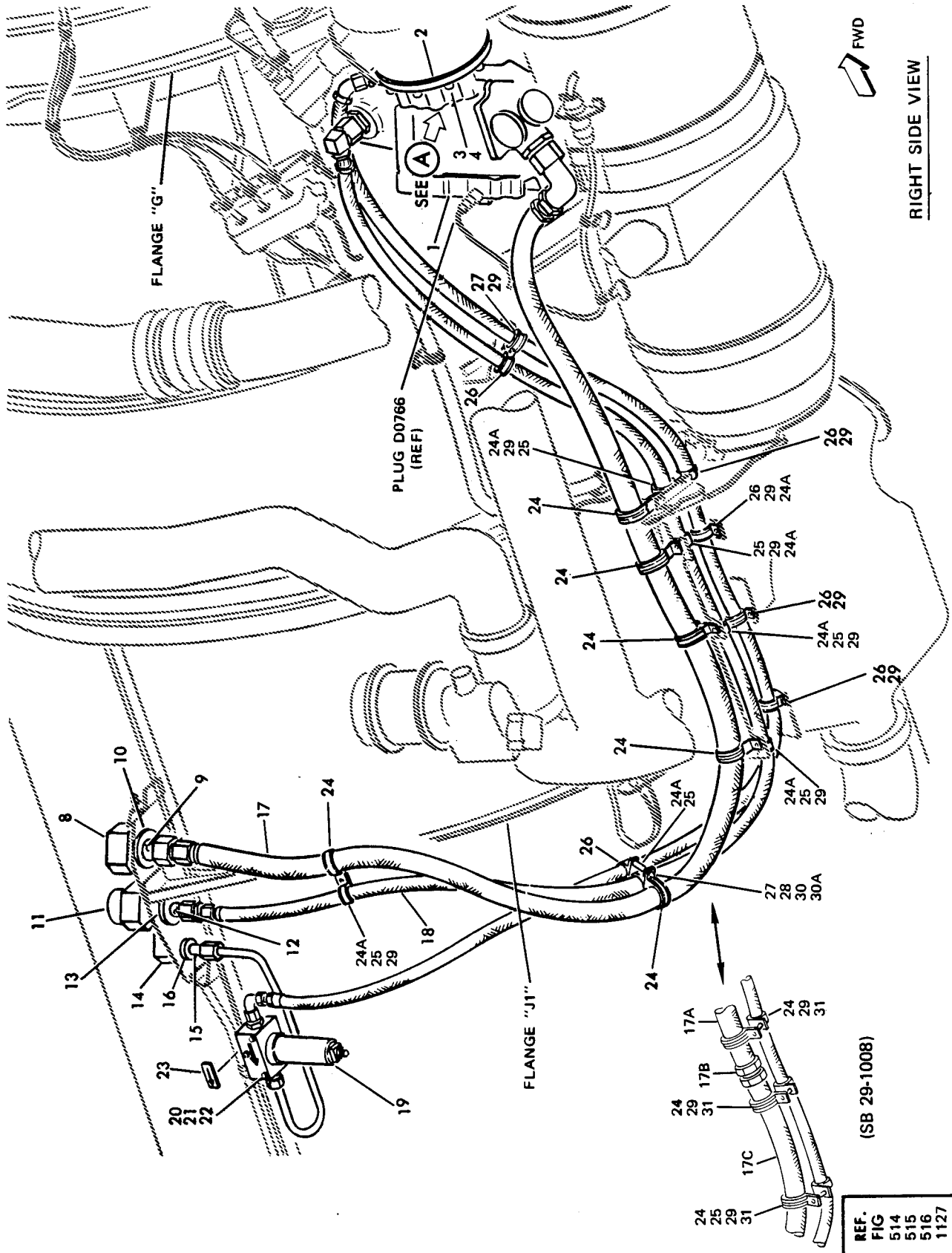
CAUTION: ATTACH HOSE ASSEMBLY TO ENGINE WITH CLAMPS BEFORE ATTACHING END FITTINGS TO PREVENT FOLDING AND COLLAPSING HOSE DURING ASSEMBLY.

(8) Install hose assy (17), or hose assy (17A), union (17B), and tube assy (17C) between coupling half (9) and hydraulic pump assy (1).

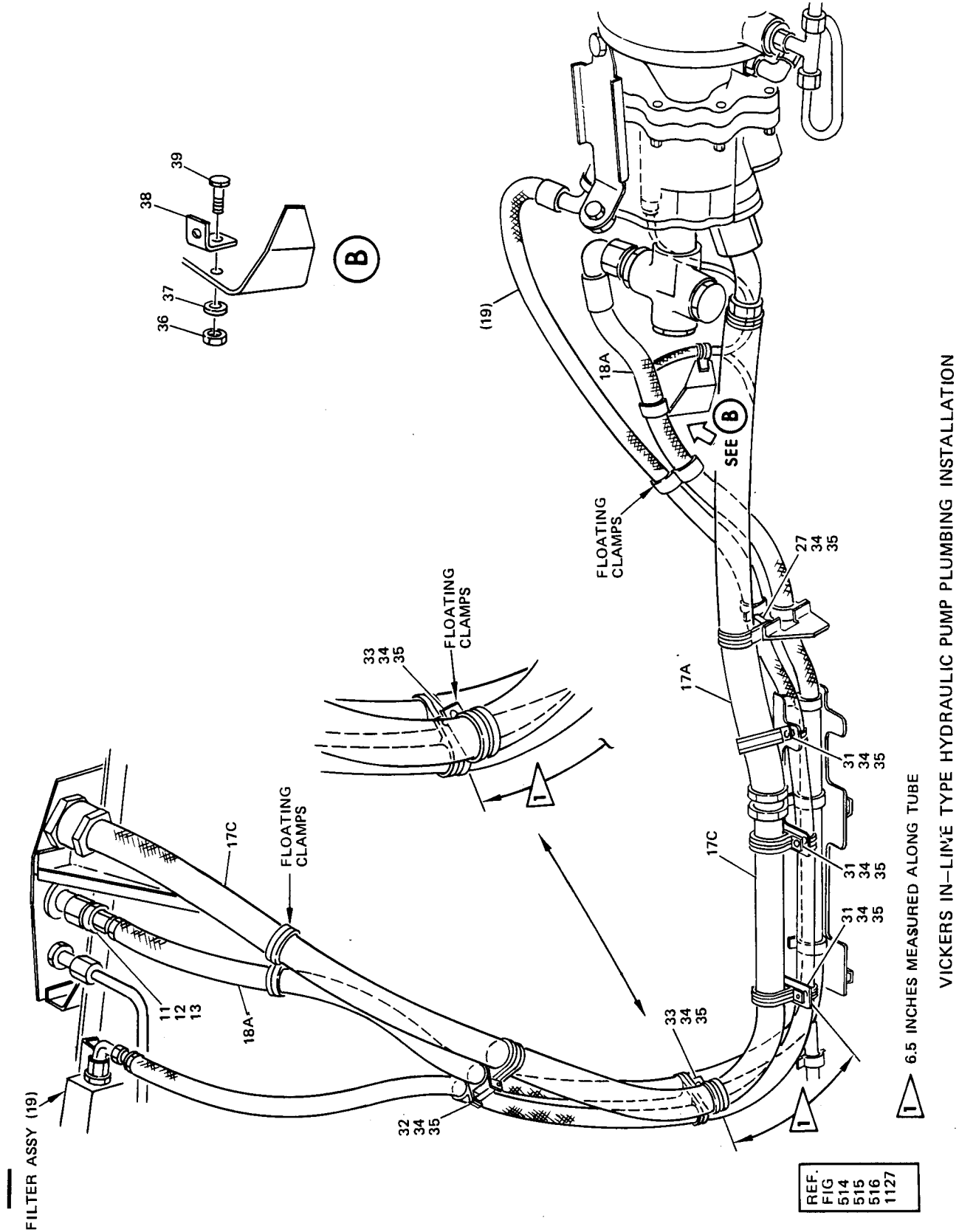
(9) Install hose (18, 18A) between coupling half (12) and pump. Tighten end fittings of hose.

(10) Locate filter assembly (19) on existing bracket on horizontal flange aft of flange J. Attach filter assembly with bolts (20), washers (21), and nuts (22). Install filter with directional arrow pointing aft.

- (11) Attach tube assembly from union in filter assembly (19) to coupling half (15). Tighten end fitting of tube.
- (12) Attach hose from fitting in IN port of filter to hydraulic pump. Tighten end fittings of hose.
- (13) Install marker (23) per 20-50-05.
- (14) Install clamps (24, 24A, 25, 26, 35) and spacers (27, 31, 32, 33) with screws (28, 29, 34), washer (30A), and nut (30).
- (15) Attach electrical plug D0766 to connector on hydraulic pump. Lockwire connector.

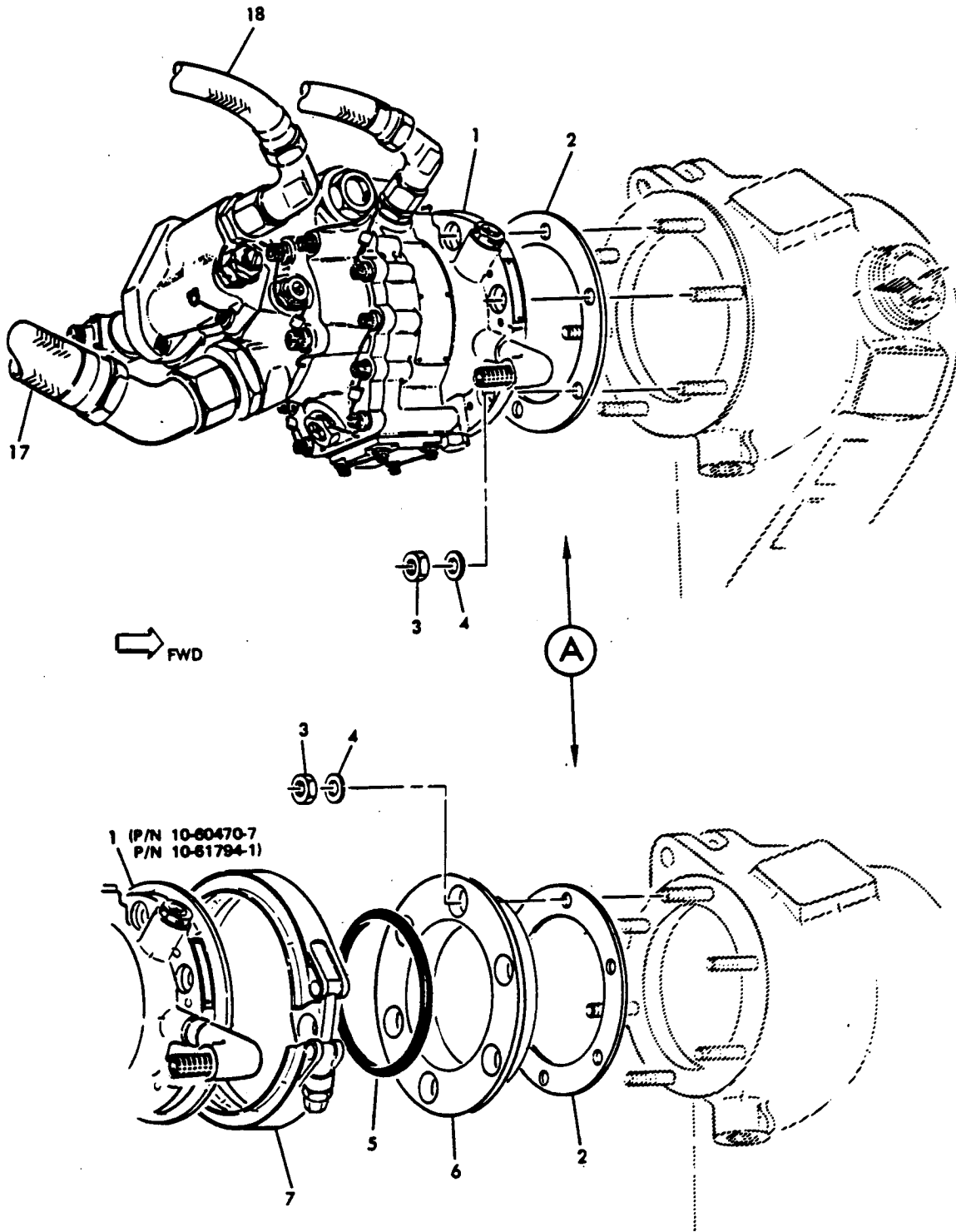


Hydraulic System Components Installation
Figure 530 (Sheet 1)



VICKERS IN-LINE TYPE HYDRAULIC PUMP PLUMBING INSTALLATION

Hydraulic System Components Installation
Figure 530 (Sheet 2)



Hydraulic System Components Installation
Figure 530 (Sheet 3)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
K.					
(1)	1	PUMP ASSEMBLY	(PREASSEMBLED IN PAR. 3.M.)		REF
(2)	2	GASKET	AN4047-1 (SUPSD BY ST1542-01)		1
(3)(a)	3	NUT	NAS679A6		6
	4	WASHER	AN960D616		6
	5	PACKING *[3]	5250612 (REPLD BY 8023812)		1
	5	PACKING *[3]	8023812 (REPLS 5250612)		1
	5	PACKING *[2]	320517		1
	6	ADAPTER *[3]	55745		1
	6	ADAPTER *[5]	382624		1
(b)	7	COUPLING *[3]	55759 (OPT TO 22807)		1
	7	CLAMP *[3]	22807 (PREF)		1
	7	MOUNTING CLAMP *[2]	374105		1
(c)		REF ITEMS 3 AND 4			
(4)	8	COUPLING HALF	015067S2-16		1
	9	COUPLING HALF	015067S5-16		1
	10	NUT	AN924-16		1
(5)	11	COUPLING HALF	307002S2-8		1
	12	COUPLING HALF	307002S5-8		1
	13	NUT	AN924-8		1
(6)	14	COUPLING HALF	307002S2-6		1
	15	COUPLING HALF	307002S5-6		1
	16	NUT	AN924-6		1
(7)		NO PARTS REQUIRED			
(8)	17	HOSE	10-60563-10		1
	17	HOSE	10-60563-19		1
(8A)	17A	HOSE	10-60563-33		1
	17A	HOSE *[4]	10-60563-34		1
	17A	HOSE	10-60563-35		1
	17B	UNION	MS21902-16J		1
	17C	TUBE ASSY	65-58145-8		1
(9)	18	HOSE	10-60563-13		1
	18	HOSE	10-60563-20		1
	18A	HOSE ASSY	BACH6R0665DU-W		1

Hydraulic System Components Installation
Figure 530 (Sheet 4)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(10)	19	FILTER ASSEMBLY	(PREASSEMBLED IN PAR. 3.M.)		REF
	20	BOLT	NAS1104-30		3
	21	WASHER	AN960D416		3
	22	NUT	BACN10JC4		3
(11)		NO PARTS REQUIRED			
(12)		NO PARTS REQUIRED			
(13)	23	MARKER	BACM10F11-3AH		1
(14)	24	CLAMP	BACC10HS21		6
	24A	CLAMP	BACC10HS17		7
	25	CLAMP	BACC10HS14		7
	26	CLAMP	BACC10HS12		6
	27	SPACER	NAS43HT3-56		1
	28	SCREW	BACCS12CB3-16		1
	29	SCREW	BACCS12CB3-7		9
	30	NUT	NAS679A3W		1
	30A	WASHER	AN960D10		7
	31	SPACER	NAS43HT3-32		3
	32	SPACER	NAS43HT3-72		1
	33	SPACER	NAS43HT3-24		1
	34	SCREW *[1]	NAS1801-3-(-)		AR
	35	CLAMP *[1]	BACC10HS-(-)		AR
	36	NUT	BACN10JC3		1
	37	WASHER	AN960-10		1
	38	BRACKET	AN743-12		1
	39	SCREW	BACCS12CB3-8		1
(15)		NO PARTS REQUIRED			

*[1] SIZE DETERMINED ON INSTALLATION

*[2] PART OF VICKERS PUMP ASSEMBLY

*[3] USED WITH ABEX PUMP ASSEMBLY

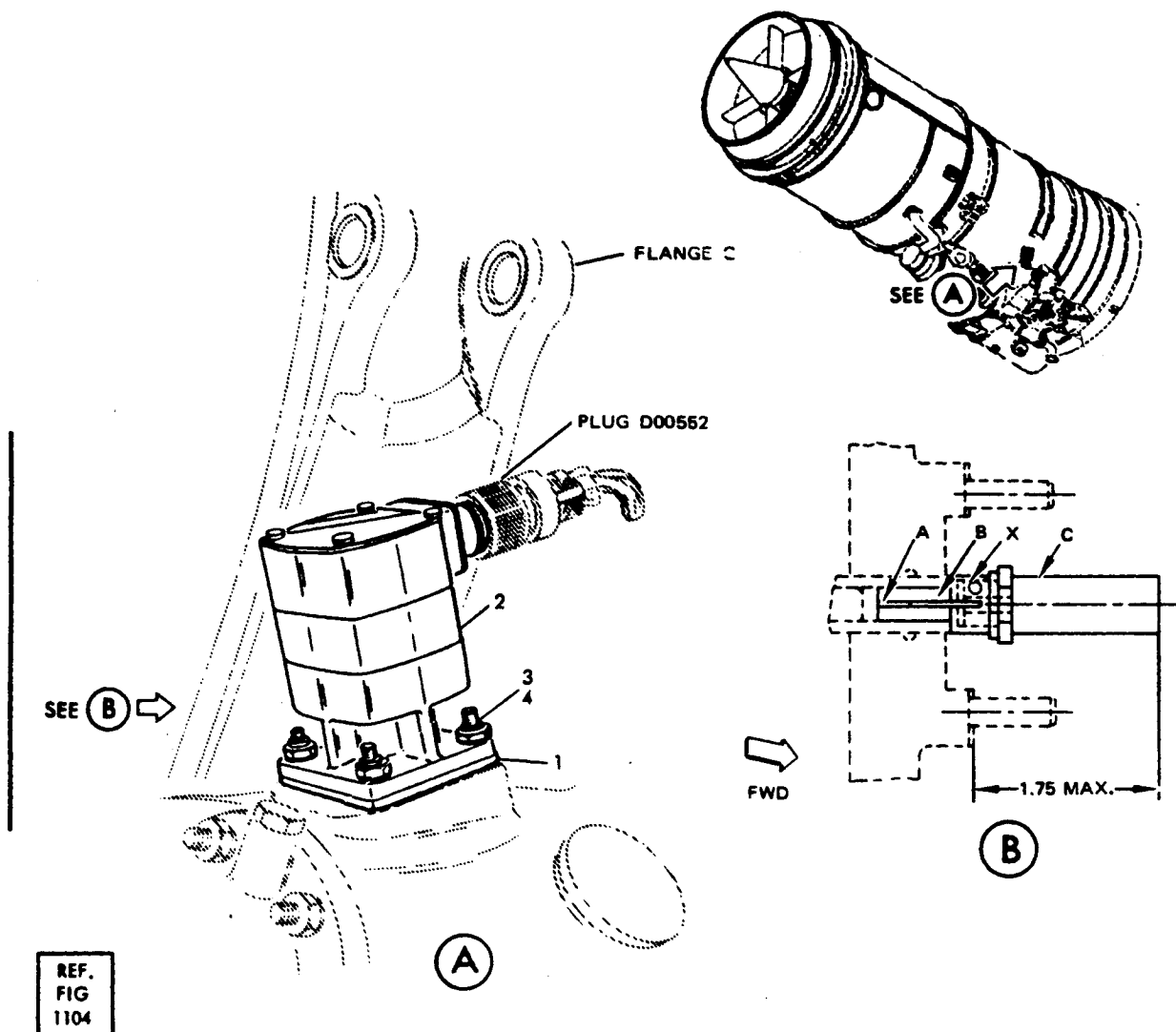
*[4] USED WITH QADMOUNTS (ABEX AND VICKERS)

*[5] PART OF VICKERS PUMP ASSEMBLY. ADAPTER NOT PROCURABLE ALONE. ORDER ADAPTER AND PINS SUBASSEMBLY, P/N 387999

Hydraulic System Components Installation
 Figure 530 (Sheet 5)

L. Install N2 tachometer generator as follows (Fig. 531):

- (1) Place gasket (1) over studs on N2 tachometer mounting pad.
- (2) Tachometer Generator (2)
 - (a) B5069110302-0 or 2CM9ABY7 -- Coat drive shaft of tachometer generator with grease MIL-G-81322 per 20-50-07.
 - (b) 2CM15AAB1 -- Before installation perform following (see **B**):
 - 1) Visually inspect surface X of rotor part B and C for any particles or other contamination that might interfere with alignment.
 - 2) Clean if necessary and relubricate surfaces X with lubricant per MIL-A-907D.
 - 3) Insert square end of rotor assy into engine shaft and tighten part C finger-tight.
 - 4) Tap end of rotor C gently with brass or plastic hammer to assure seating with engine shaft, and tighten rotor C to 40 lb-in.
- NOTE: Make electrical bond between attaching hardware and tachometer generator flange. Maximum allowable resistance between generator and basic structure is 0.0025 ohms (2.5 milliohms).
- (3) Attach tachometer generator with washers (3) and nuts (4).
- (4) Attach plug D0552 to electrical connector on tachometer generator, and lockwire in place.



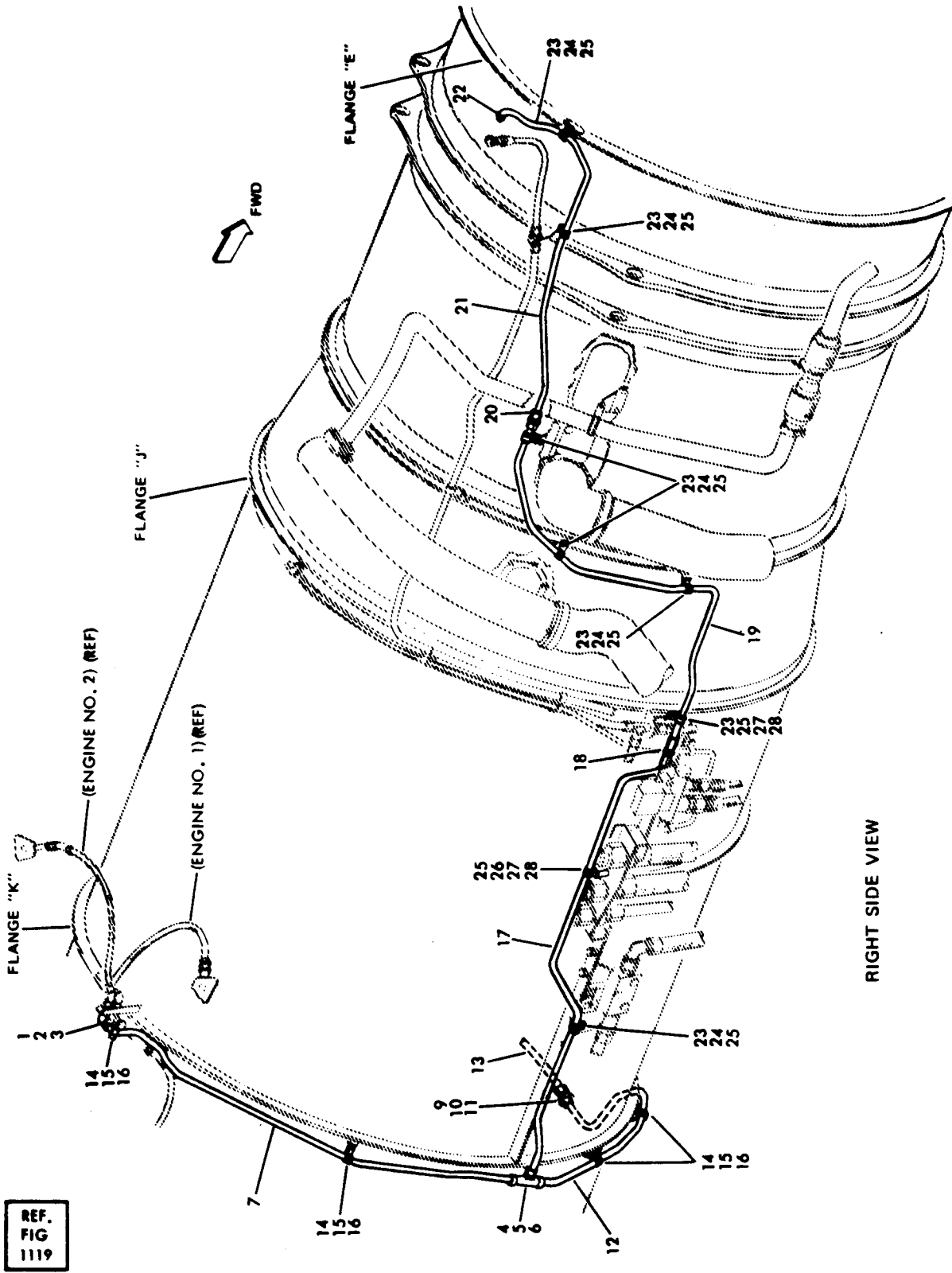
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
L. (1)	1	GASKET	AN4045-1		1
(2)	2	TACHOMETER GENERATOR	B5069110302-0		1
	2	TACHOMETER GENERATOR	2CM9ABY7		1
	2	TACHOMETER GENERATOR	2CM15AAB1		1
(3)	3	WASHER	AN960PD416L		4
	4	NUT	NAS679A4W		4
(4)		NO PARTS REQUIRED			

M. Install strut drain tubing as follows: (See figure 532.)

NOTE: This paragraph applies to engines with pneumatically-operated thrust reversers.

- (1) Install elbow (1), washer (2), and nut (3) on existing bracket at top of flange "K".
- (2) Install tee (4), washer (5), and nut (6) on existing bracket on right side of flange "K".
- (3) Install tube assembly (7) between elbow (1) and tee (4). Tighten end fittings of tube assembly.
- (4) Deleted.
- (5) Install union (9), washer (10), and nut (11) on existing bracket on lower right side of aft mounting flange. Tighten nut (11).
- (6) Install tube assembly (12) between union (9) and tee (4). Tighten end fittings of tube assembly.
- (7) Attach hose (13) to bottom port of union (9). Stow free end of hose. (Refer to paragraph AL.)
- (8) Secure tube assemblies with clamps (14), screws (15). Use nuts (16) where nutplates are not installed on support brackets.
- (9) Attach tube assembly (17) to tee (4). Tighten end fitting of tube assembly.
- (10) Install union (18) on forward end of tube assembly (17). Tighten end fitting of tube assembly.
- (11) Attach tube assembly (19) to union (18). Tighten end fitting of tube assembly.
- (12) Install union (20) on forward end of tube assembly (19). Tighten end fitting of tube assembly.
- (13) Attach tube assembly (21) to union (20). Tighten end fitting on tube assembly.
- (14) Install union (22) on tube assembly (21). Tighten end fitting of tube assembly.
- (15) Secure tube assemblies (17, 19, and 21) to existing brackets with seven clamps (23) and screws (24). Use nuts (25) where nutplates are not installed on support brackets. Install clamp (26), spacer (27), screw (28), and nut (25).
- (16) Cap open ports of union (22) and hoses.

BOEING
COMMERCIAL JET
OVERHAUL MANUAL



REF.
 FIG.
 1119

Strut Drain Tubing Installation
 Figure 532 (Sheet 1)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
M.					
(1)	1	ELBOW	MS21908-6		1
	2	WASHER	AN960-916		1
	3	NUT	AN924-6		1
(2)	4	TEE	MS21909-6		1
	5	WASHER	AN960-916		1
	6	NUT	AN924-6		1
(3)	7	TUBE ASSEMBLY	65-55475-14		1
(4)	8	DELETED	10-60846-7		
(5)	9	UNION	MS21924-6		1
	10	WASHER	AN960-916		1
	11	NUT	AN924-6		1
(6)	12	TUBE ASSEMBLY	65-55475-13		1
(7)	13	HOSE	BACH30AV06W0163		1
(8)	14	CLAMP	BACC10HS06		4
	15	SCREW	BACS12CB3-7		4
	16	NUT	NAS679A3W		AR
(9)	17	TUBE ASSEMBLY	65-55475-803		1
(10)	18	UNION	MS21902-6		1
(11)	19	TUBE ASSEMBLY	65-55475-802		1
(12)	20	UNION	MS21902-6		1
(13)	21	TUBE ASSEMBLY	65-55475-16		1
(14)	22	UNION	MS21924-6		1
(15)	23	CLAMP	BACC10HS06		6
	24	SCREW	BACS12CB3-7		7
	25	NUT	NAS679A3W		AR
	26	CLAMP	BACC10HS06		1
	27	SPACER	NAS43HT3-40		2
	28	SCREW	BACS12CB3-20		2
(16)		NO PARTS REQUIRED			

N. Install airconditioning sensing plumbing as follows: (Fig. 533)

- (1) Install union (1) on existing bracket on flange F, just inboard of right forward engine mount. Fasten union with washer (2) and nut (3). Tighten nut.
- (2) Install tube assembly (4) between fitting in top forward port of pressure differential regulator and union (1). Connect A/C sense line hose assembly (4A) to union (1). Tighten end fittings of both assemblies.

NOTE: Hose assembly (4A) is not part of the QEC and normally remains attached to the strut. It is included here to illustrate the configuration found if an operator chooses to disconnect the hose assembly at the strut during engine removal.

- (3) Install elbow (5 or 5A) with washer (6) and nut (7) on existing bracket on left bleed port or flange H. Do not tighten nut.
- (4) Install tube assembly (8 or 8A) between elbow (5 or 5A) and elbow in aft center port of pressure differential regulator. Tighten end fittings of tube assembly. Tighten nut (7).
- (5) Install hose (9, 9A, or 9B) between elbow (5, 5A, or 5B) and fitting in boss in engine starter manifold. Tighten end fittings of hose.

NOTE: When replacing hose 66081 (9A) with preferred configuration, 65-58126-23 (9A), 450-degree F thermostat must be removed, rotated 90 degrees, and reinstalled to accommodate rerouting of the pneumatic sensing line.

- (6) Install elbow (10) in existing bracket on left side of flange J at horizontal centerline of engine. Attach elbow with washer (11) and nut (12). Do not tighten nut.
- (7) Install tees (13 and 15A) just below elbows (10), in same bracket, with washers (14) and nuts (15). Do not tighten nuts. Install relief valve (15B) with packing (15C) in end of tee (13).

NOTE: When optional elbow (13) is installed use one washer (14) and one nut (15). Do not tighten nut.

- (8) Install hose (16 or 16A) between elbow (10) and fitting in end of high pressure air manifold. Tighten end fittings of hose. Tighten nut (12).
- (9) Install tube assembly (17) between elbow (13) and tee on flange J, just forward of pressure differential regulator. Tighten end fittings of tube assembly.

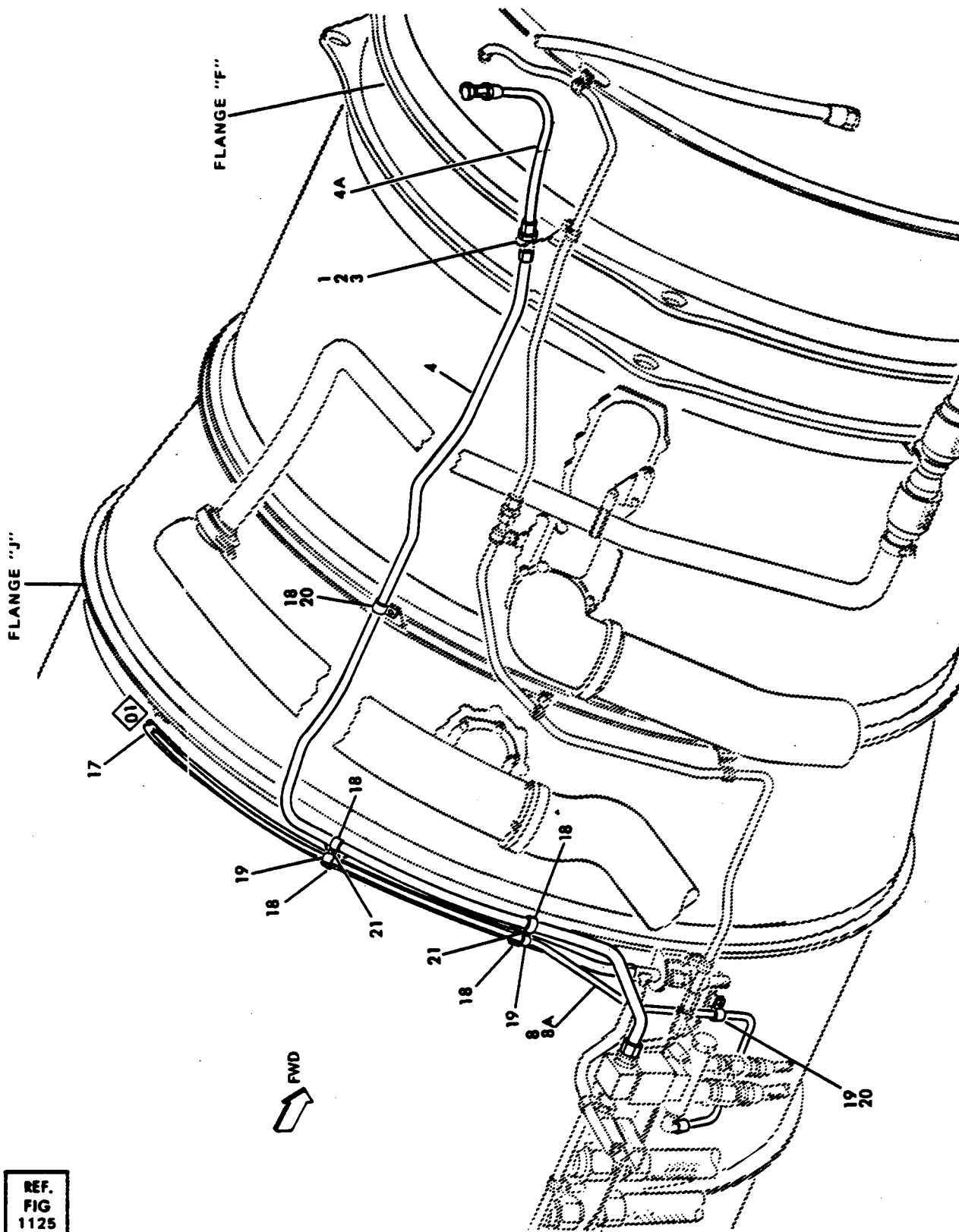
737

BOEING 
COMMERCIAL JET

POWERPLANT BUILDUP

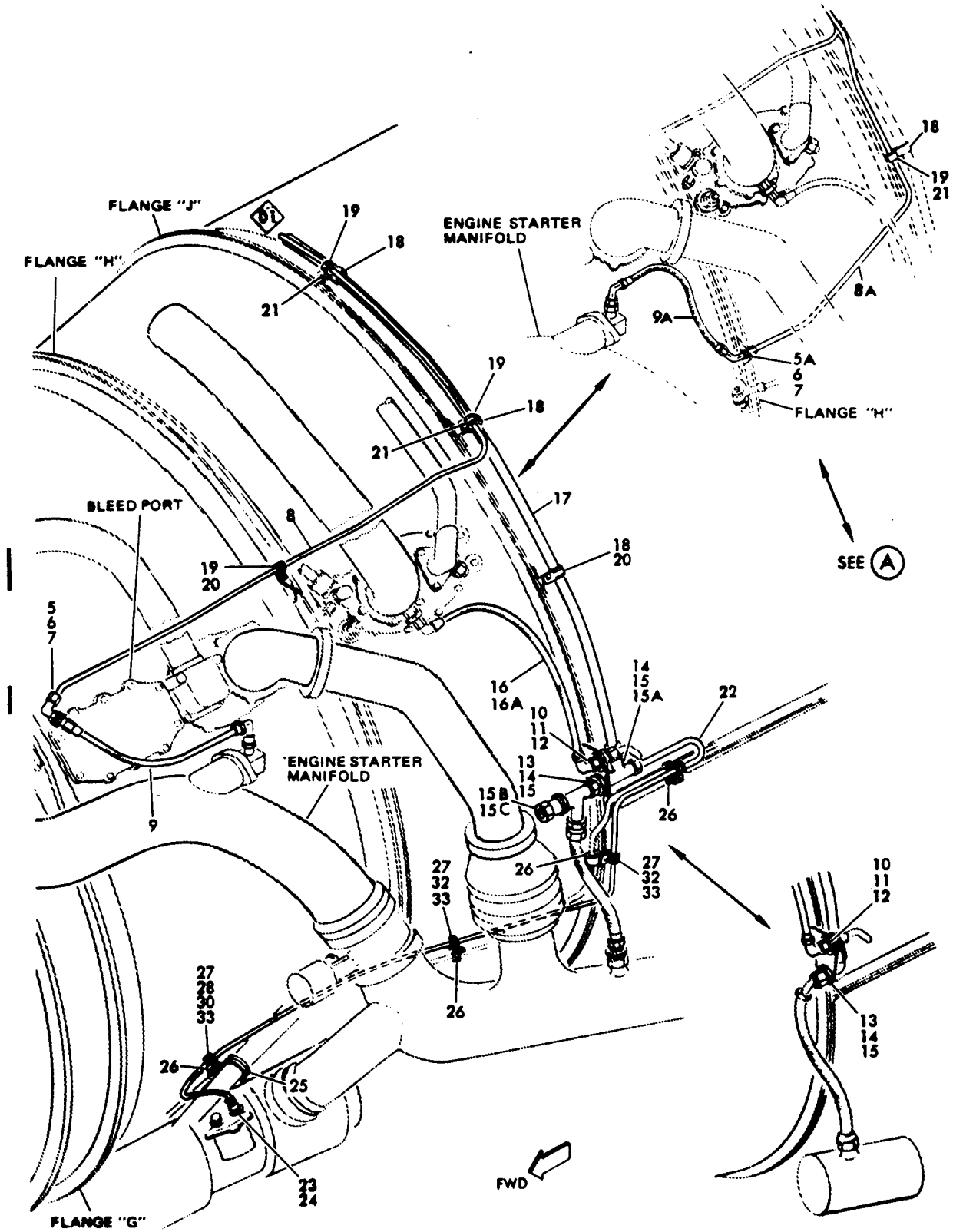
OVERHAUL MANUAL

- (10) Locate clamps (18) and (19). Secure clamps with screws (20 and 21).
- (11) Deleted.
- (12) Install tube assembly (22) and filter assembly (23), between tee (15A) and aft port on starter assembly, with packing (24). Tighten end fittings of tube assembly.
- (13) Install clamps (25, 26 and 27) and spacers (28 and 29) with screws (30, 31 and 32) and nuts (33).
- (14) Cap open ports in lines.

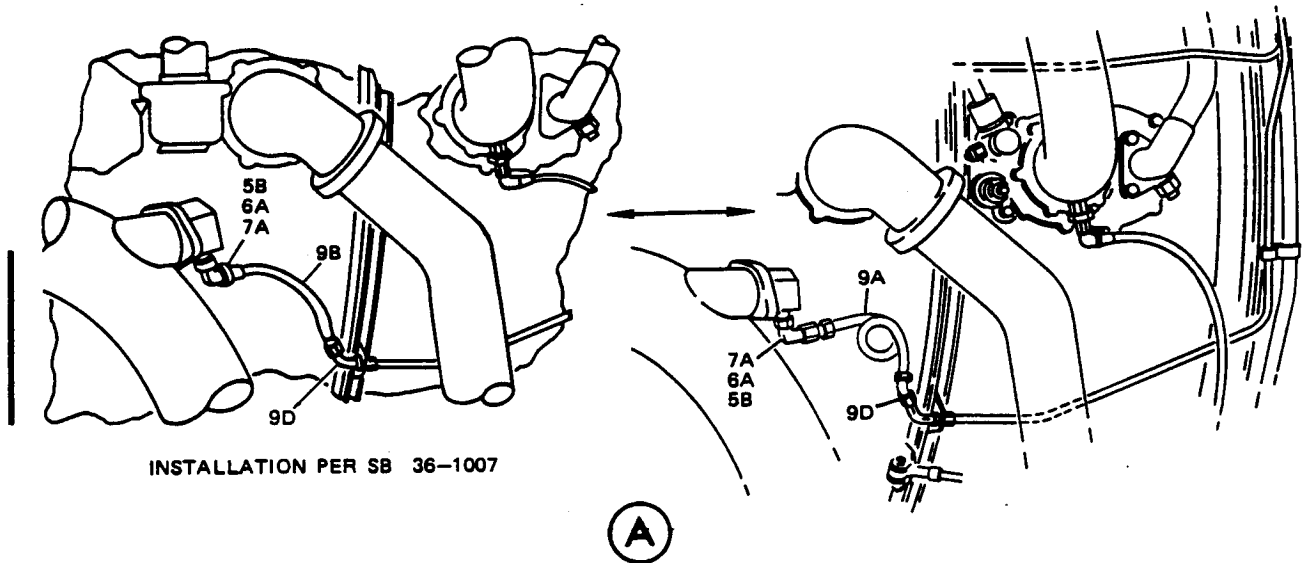


REF.
FIG
1125
1127

Air-Conditioning Sensing Plumbing
Figure 533 (Sheet 1)



Air-Conditioning Sensing Plumbing
 Figure 533 (Sheet 2)



Air Conditioning Sensing Plumbing
Figure 533 (Sheet 3)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
N. (1)	1	UNION	MS21924-6		1
	2	WASHER			
(2)	3	NUT	AN960-916		1
	4	TUBE ASSEMBLY	AN924-6		1
	4A	HOSE ASSEMBLY	65-58126-10		1
	4A	HOSE ASSEMBLY	AS136-06-0131 *[7]		1
(3)	4A	HOSE ASSEMBLY	AS137-06-0146 *[6]		1
	4A	HOSE ASSEMBLY	16235 *[8]		1
	5	ELBOW	MS21908-4 *[2]		1
	5A	ELBOW	MS21907-4 *[4]		1
	5B	ELBOW	MS21926-4 *[6]		1
	6	WASHER	AN960-716		1
	6A	GASKET	AN901-4C *[6]		1
(4)	7	NUT	AN924-4		1
	7A	NUT	AN924-4 *[6]		1
	8	TUBE ASSEMBLY	65-58126-800 *[2]		1
	8A	TUBE ASSEMBLY	65-58126-21 *[4]		1
(5)	9	HOSE	AS138-04-0084 *[6]		1
	9	HOSE	16233 (AVICA)		1
	9A	HOSE *[6] (PREF)	65-58126-23		1
	9A	HOSE	66081 (BELLOW)*[5]		1
	9B	HOSE *[6] (PREF)	65-58126-23		1
	9B	HOSE	109339-2 *[6]		1
	9D	ELBOW	BACE21AT0406 *[6]		1
	9D	ELBOW *[6] (PREF)	MS21907J4		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(6)	10	ELBOW	MS21908-4 *[3]		1
	11	WASHER	AN960-716		1
	12	NUT	AN924-4		1
(7)	13	TEE	118-10430		1
	13	ELBOW	MS21908-6 *[1]		1
	14	WASHER	AN960-916		2
	15	NUT	AN924-6		2
	15A	TEE	BACT16AX060406		1
	15B	RELIEF VALVE	1667		1
	15C	PACKING	BACP11PB8		1
(8)	16	HOSE	16231 (AVICA)		1
	16A	HOSE	AS138-04-0132 *[6]		1
(9)	17	TUBE ASSEMBLY	65-58126-6		1
(10)	18	CLAMP	BACC10HS06		8
	19	CLAMP	BACC10HS04		7
	20	SCREW	BACS12CB3-7		4
	21	SCREW	BACS12CB3-9		6
(11)		DELETED			
(12)	22	TUBE ASSY	65-58126-12		1
	23	FILTER ASSY	11-10865		1
	24	PACKING	BACP11PB6		1
(13)	25	CLAMP	BACC10HS20		1
	26	CLAMP	BACC10HS06		4
	27	CLAMP	BACC10HS04		4
	28	SPACER	NAS43HT3-24		1
	29	SPACER	NAS43HT3-16		1
	30	SCREW	BACS12CB3-16		1
	31	SCREW	BACS12CB3-12		1
	32	SCREW	BACS12CB3-7		2
	33	NUT	NAS679A3W		2

- *[1] OPTIONAL PART
- *[2] USE WITH 16233
- *[3] USE WITH 65-58126-800
- *[4] USE WITH 66081
- *[5] USE WITH 65-58126-21
- *[6] POST SB 36-1007
- *[7] PREFERRED HOSE ASSEMBLY ON ENGINE NO. 1 WHEN POTABLE WATER PRESSURE INSTALLATION IS USED
- *[8] PRE SB 36-1007

- O. Install nose cowl anti-icing system and vortex dissipator plumbing as follows (Fig. 534):

NOTE: To avoid preload, loosen all ducts and brackets when installing thermostatic valve assembly.

- (1) For engines without vortex dissipator:

- (a) Locate preassembled thermostatic valve assembly (1) bracket on flange D. If bracket 65-45799-129 or -141 is used, remove reducer and place end of valve through hole in bracket, using one gasket on each side of bracket, and reinstall and tighten reducer. Attach mounting flange of valve to bracket on engine flange D with two bolts, washers, and nuts which were loosely installed on valve assembly.

NOTE: Make electrical bond between structure and valve at attaching points. Maximum allowable resistance is 0.0017 ohm.

- (b) Duct Assembly Installation

For duct assembly (4B), attach preassembled thermostatic valve assembly (1) to bracket part number 65-45799-174 (64, IPL Fig. 1109) on flange "G" with clamp (2A), screw (2B) and nut (3A).

For duct assembly (4C), attach preassembled thermostatic valve assembly (1) to bracket part number 65-45799-52 (64, IPL Fig. 1109) on flange "G" with clamp (2) and nut (3).

- (c) Attach duct assembly (4, 4A, 4B) to port in air conditioning duct (saddle duct) at top centerline of engine with clamp (6). Install clamp with bolt at 180° (bottom) position.
- (d) Attach duct assembly (4, 4A, 4B) to preassembled thermostatic valve assembly (1) with clamp (5).
- (e) Install duct assembly (7) on forward end of thermostatic valve assembly. Do not tighten nut on duct assembly.
- (f) Install clamps (8) with screws (9) and nuts (10).

NOTE: Shim between indicated clamp (8) and support bracket with washers (9A) as required to prevent preload on TAI duct (4 or 4A). Limited to maximum 0.30 inch shim buildup.

- (g) Tighten end nut on duct assembly (7) and attach ring (11) on duct assembly for connection with nose cowl.
- (h) Attach electrical plug D0196 to connector on thermostatic valve. Lockwire plug in place.
- (i) Plug open ports in filter and duct assembly.

- (2) For engines with boom type vortex dissipator:

- (a) Locate thermostatic valve (12) on existing bracket on flange D. Attach with two bolts (13), washers (14), and nuts (15) which were preassembled with valve.

NOTE: Make electrical bond between structure and valve at attaching points. Maximum allowable resistance is 0.0017 ohm.

- (b) Install duct assembly (16), attaching with clamps (17), screws (18), and nuts (19) in two places and with clamp (20) secured to fuel line at flange C by means of clamp (21), spacer (22), screw (23) and nut (24).
- (c) Connect aft end of duct assembly (16) to TAI ducting with coupling (25).
- (d) Position gasket (26) on upper flange of thermostatic valve (12) and attach flange of duct assembly (16) with four bolts (27) or two bolts (27A) and two bolts (27B) with washers (28), and nuts (29).
- (e) Attach forward end of duct assembly (16) to bracket at flange B with clamp (30), screw (31), washer (32), and nut (33).
- (f) Install duct assembly (34) on thermostatic valve (12) and secure to existing brackets on flanges C and B with clamps (35), screws (36), and nuts (37).
- (g) Tighten end nut on duct assembly (34) and attach ring (38) for connection with nose cowl.
- (h) Install valve (39) on duct assembly (16) with gasket (40) and coupling (41). Ensure valve is 1.25-1.45 inch from engine case.
- (i) Install duct assembly (42) on valve (39) with gasket (43) and coupling (44), and secure duct assembly with clamp (45), screw (46), washer (47), and nut (48).
- (j) Tighten end nut on duct assembly (42) and attach ring (49) for connection with nose cowl.
- (k) Install union (50) with O-ring gasket (51) in port on valve (39).
- (l) On duct assembly (42), at first boss from valve (39), install tee (52) with O-ring gasket (53) and nut (54). Install cap (55) on tee.

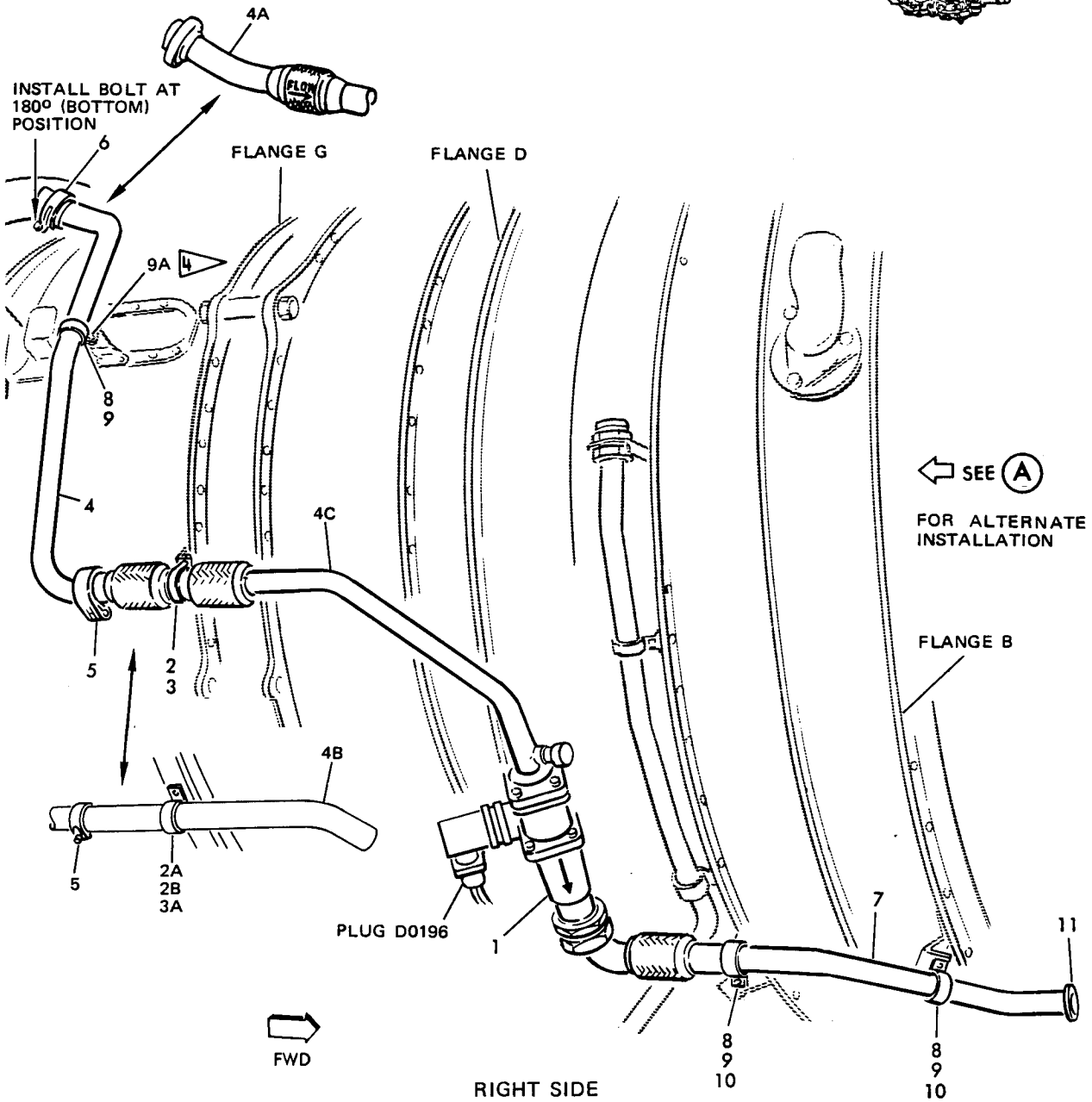
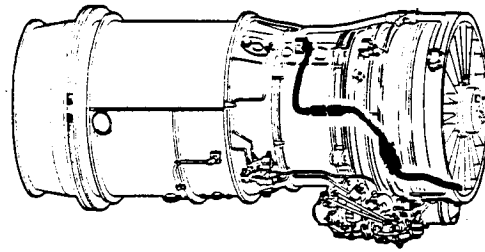
OVERHAUL MANUAL

NOTE: On 65-58126-14 installations not electing to use bracket mounted switch, omit steps (m), (n) and (o) and perform steps (p) and (q).

- (m) Install pressure switch (56) on bracket on flange B with clamp (57) and screw (58).
- (n) On duct assembly (42), at second boss from valve (39), install union (59) with O-ring gasket (60), and install tube assembly (61) between it and switch (56).
- (o) Install tube assembly (62) between tee (52) and union (50) in valve (39).

NOTE: Steps (p) and (q) apply only to 65-58126-14 installations not electing to use the bracket mounted switch.

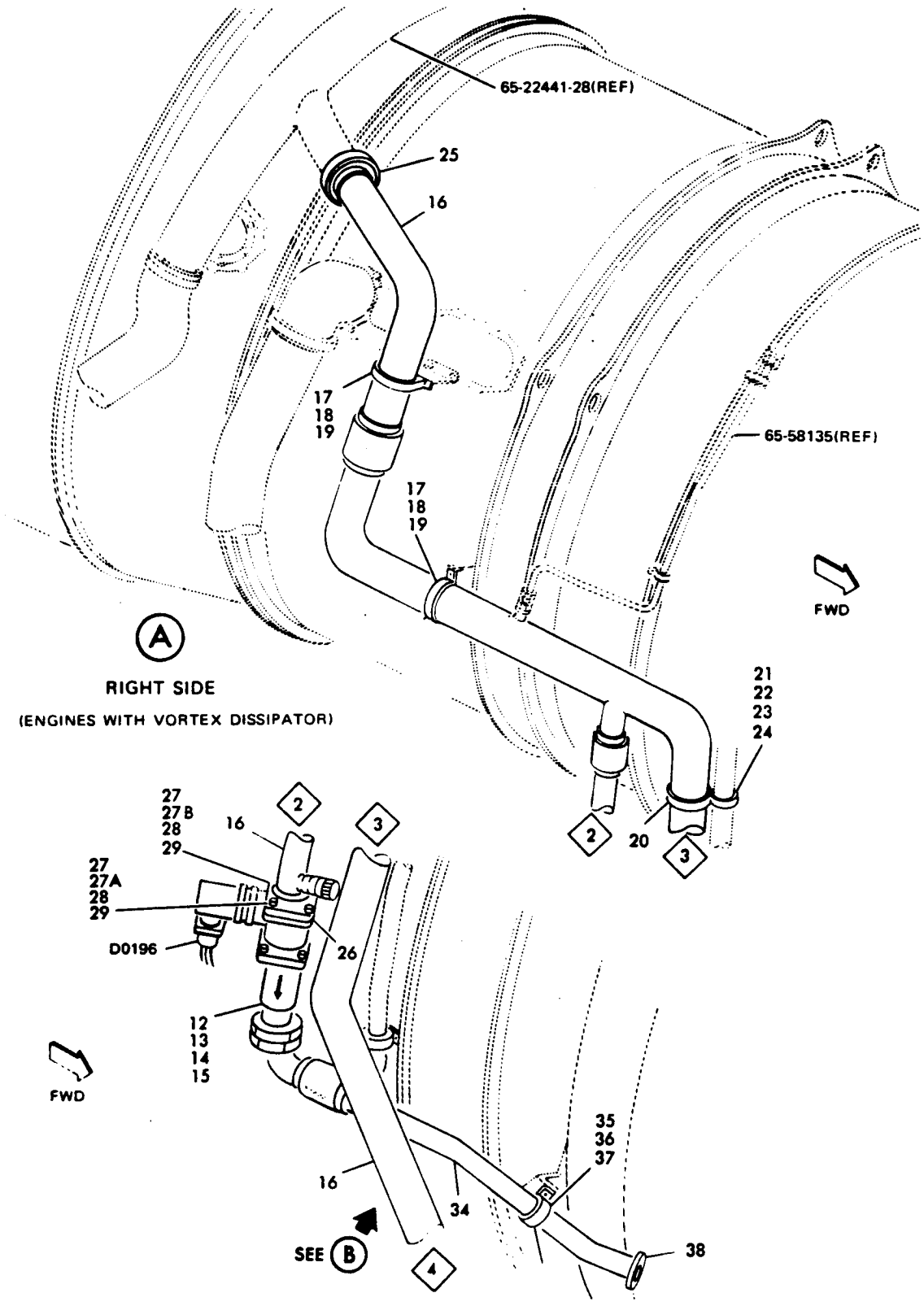
- (p) On duct assembly (42), at second boss from valve (39), install pressure switch (64) with O-ring gasket (65).
- (q) Install tube assembly (66) between tee (52) and union (50) in valve (39).
- (r) Attach electrical plug D0196 at connector on thermostatic valve (12).
- (s) Check that all flanges and brackets are tight.



REF.
FIG
1125

INSTALL SHIM (9A) AT THIS POSITION ONLY IF NECESSARY TO PREVENT PRELOAD CONDITIONS ON TAI DUCT (4, 4A).

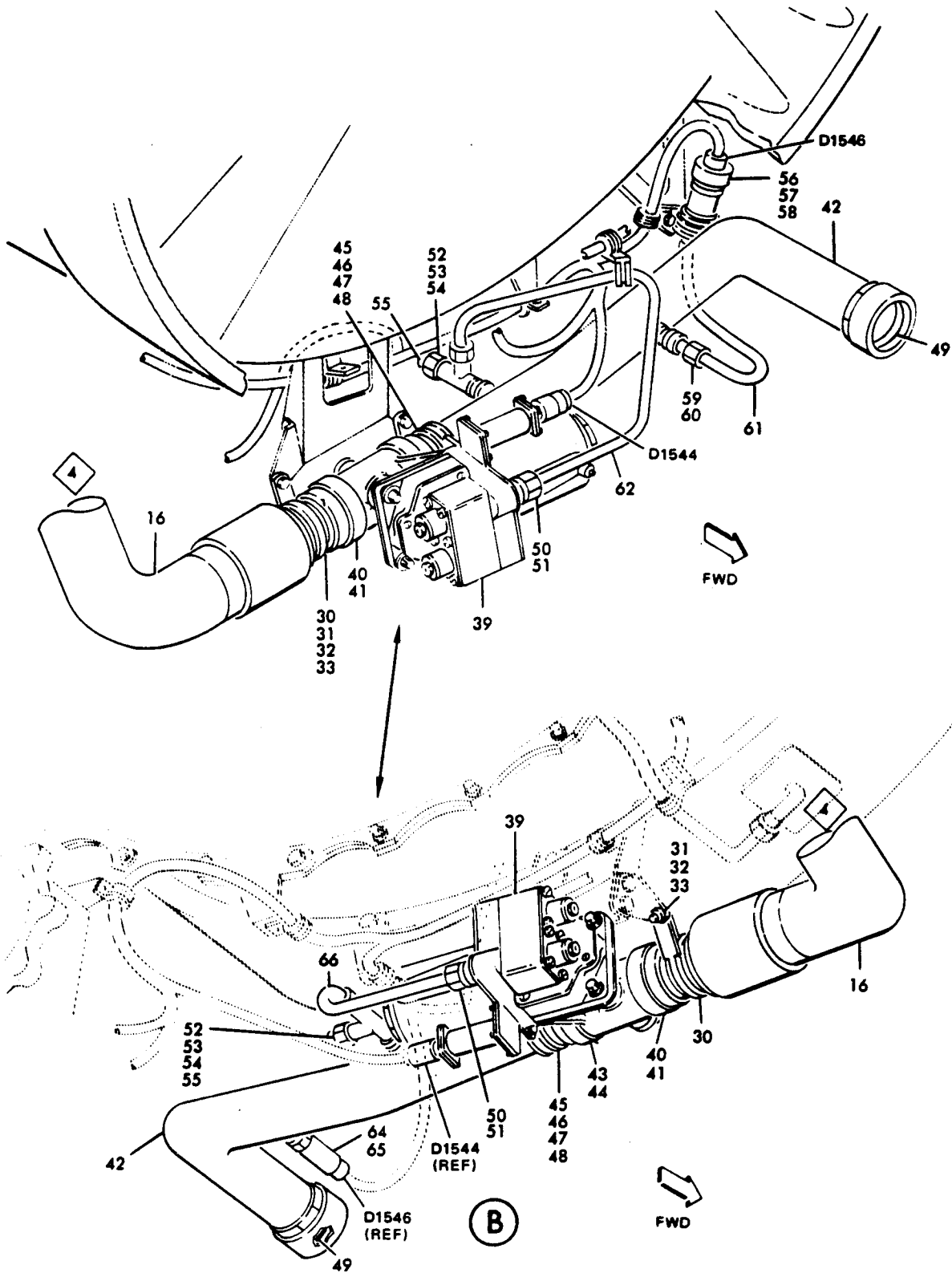
Nose Cowl Anti-Icing Components Installation
Figure 534 (Sheet 1)



(A)
RIGHT SIDE
(ENGINES WITH VORTEX DISSIPATOR)

Nose Cowl Anti-Icing Components Installation
Figure 534 (Sheet 2)

Sep 10/71



Nose Cowl Anti-Icing Components Installation
 Figure 534 (Sheet 3)

Sep 10/71

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(1)(a)	1	VALVE ASSY THERMOSTATIC	(PREASSEMBLED IN PAR. 3.0.)		REF
(b)	2	CLAMP	BACC10V22-6C		1
	2A	SADDLE CLAMP	NAS1716C20M		1
	2B	SCREW	NAS1802-3-8		2
	3	NUT	NAS679A3W		2
	3A	NUT	MS21042L3		2
(c)	4	DUCT ASSY	65-58123-1		1
	4A	DUCT ASSY	65-58123-10		1
	4B	DUCT ASSY	65-58123-13		1
	4C	DUCT ASSY	65-58123-2		1
	5	CLAMP	BACC10DU125C		1
(d)	6	CLAMP	BACC10DU150C		1
(e)	7	DUCT ASSY	65-58123-7		1
(f)	8	CLAMP	BACC10HS20		3
	9	SCREW	BACS12CB3-7 *[3]		3
	9A	WASHER	AN960C10		AR
	9A	WASHER	AN960C10L		AR
	10	NUT	NAS679A3W		2
(g)	11	RING	NAS595-20E		1
(h)			NO PARTS REQUIRED		
(i)			NO PARTS REQUIRED		
(2)(a)	12	VALVE, THERMOSTATIC	(PREASSEMBLED IN PAR. 3.0)		REF
	13	BOLT	(PREASSEMBLED WITH VALVE)		REF
	14	WASHER	(PREASSEMBLED WITH VALVE)		REF
	15	NUT	(PREASSEMBLED WITH VALVE)		REF
(b)	16	DUCT ASSY	65-77452-1		1
	16	DUCT ASSY	65-77452-13		1
	17	CLAMP	BACC10CD32		2
	18	SCREW	BACS12CB3-6		4
	19	NUT	BACN10JC3		4
	20	CLAMP	BACC10HS32		1
	21	CLAMP	BACC10HSO6		1
	22	SPACER	NAS43HT3-16		1
	23	SCREW	BACS12CB3-6		1
	24	NUT	BACN10JC3		1
(c)	25	COUPLING	BACC10DU200X		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(d)	26	GASKET	60023-1 *[2]		1
	27	BOLT	AN5C10A *[1]		4
	27A	BOLT	BACB30LM5U10		2
	27B	BOLT	BACB30LM5U12		2
	28	WASHER	AN960C516		4
	29	NUT	NAS679C5		4
(e)	30	CLAMP	BACC10HS24		1
	31	SCREW	BACSL2CB3-6		1
	32	WASHER	AN960-10L		1
	33	NUT	BACN10JC3		1
(f)	34	DUCT ASSY, TAI	65-58123-7		1
	35	CLAMP	BACC10HS20		2
	36	SCREW	BACSL2CB3-7		2
	37	NUT	NAS679A3W		2
(g)	38	RING	NAS595-20E		1
(h)	39	VALVE	F6100236M4		1
	39	VALVE	F6100236M3 (OPTL)		1
	40	GASKET	24096-150C		1
	41	COUPLING	24502-150		1
(i)	42	DUCT ASSY	65-77452-10		1
	43	GASKET	24096-150C		1
	44	COUPLING	24502-150		1
	45	CLAMP	BACC10HS24		1
	46	SCREW	BACSL2CB3-6		1
	47	WASHER	AN960-10L		1
	48	NUT	BACN10JC3		1
(j)	49	RING	NAS595-24E		1
(k)	50	UNION	AN21902-4C		1
	51	GASKET, O-RING	AN901-4C		1
(l)	52	TEE	MS21912-4C		1
	53	GASKET, O-RING	AN901-4C		1
	54	NUT	AN924-4J		1
	55	CAP	BACC14H4LJ		1

737

POWERPLANT BUILDUP



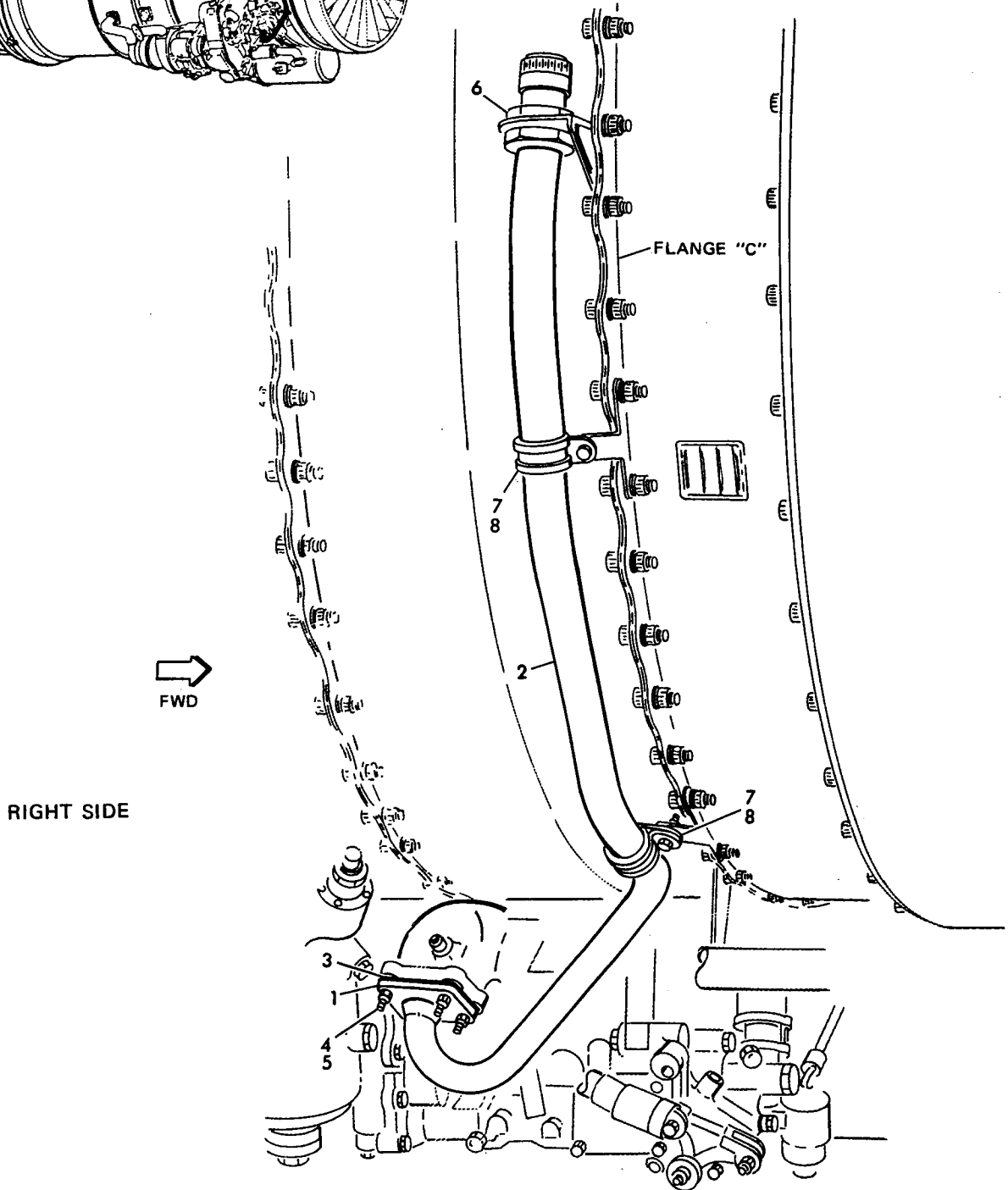
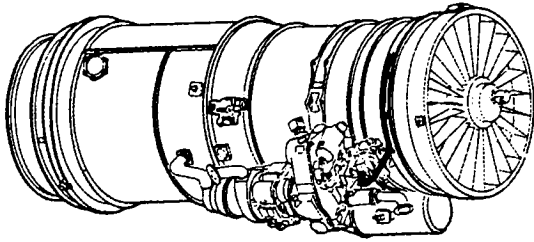
OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(m)	56	SWITCH, PRESSURE	7026 (USED ON 65-58126-20; OPTL ON 65-58126-14)		1
	57	CLAMP	BACC10HS18		1
	58	SCREW	BACS12CB3-7		1
(n)	59	UNION	AN21902-4J		1
	60	GASKET, O-RING	AN901-4C		1
	61	TUBE ASSY	65-58126-19		1
(o)	62	TUBE ASSY	65-58126-18		1
(p)	63	DELETED			
(q)	64	SWITCH, PRESSURE	8G654 (USED ON 65-58126-14 ONLY) 7026 (OPTL ON 65-58126-14)		1
	65	GASKET, O-RING	AN901-4C		1
(r)	66	TUBE ASSY	65-58126-17		1

- *[1] Optional to BACB30LM5U10 and BACB30LM5U12
- *[2] Replaces 60023
- *[3] Increase in screw length may be necessary to accommodate shim (9A) installation

P. Install fuel inlet tube as follows: (See Fig. 535.)

- (1) Slide flange (1) over end of tube assembly (2). Seat flange of tube in flange (1).
- (2) Place gasket (3) over studs on fuel inlet pad on lower right side of engine. Seat flange (1) on gasket and attach with four washers (4) and nuts (5).
- I (3) Lockwire all nuts (5) as shown in SOPM 20-50-02.
- (4) Attach tube assembly (2) to existing bracket on flange "C" with nut (6).
- (5) Attach tube assembly (2) to existing brackets on flange "C" with clamps (7) and screws (8).
- (6) Cap open end of tube assembly (2).



REF.
FIG
1126

Fuel Inlet Tube Installation
Figure 535 (Sheet 1)

737
POWERPLANT BUILDUP



OVERHAUL MANUAL

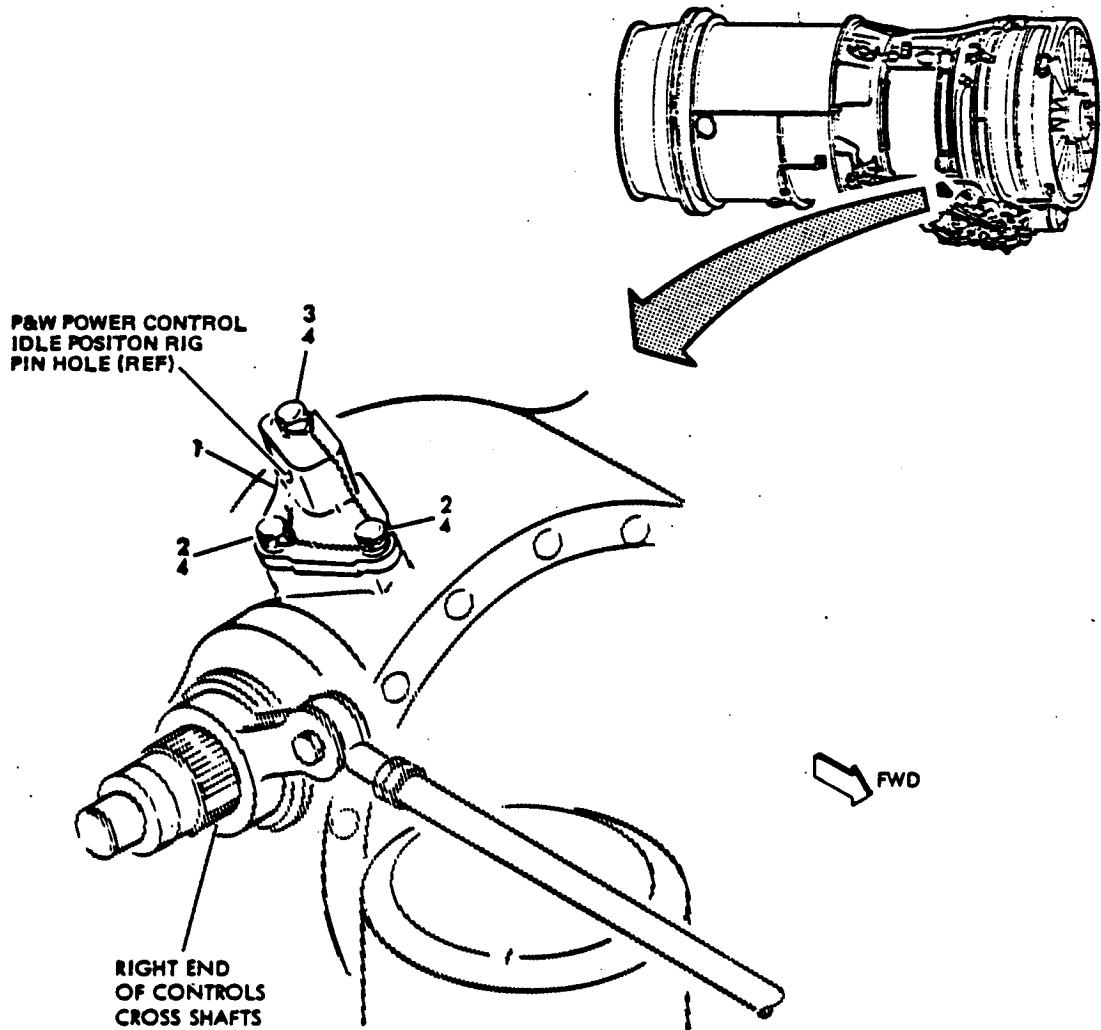
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
P. (1)	1	FLANGE	MS20756-24		1
	2	TUBE ASSEMBLY	65-46869-4		1
(2)	3	GASKET	BACG10AG305		1
	4	WASHER	AN960PD416L		4
	5	NUT	AN320-4		4
(3)		NO PARTS REQUIRED			
(4)	6	NUT	AN924-16		1
(5)	7	CLAMP	BACC10HS16		2
	8	SCREW	BACS12CB3-7		2
(6)		NO PARTS REQUIRED			

Q. Install engine controls rigging pin mount as follows: (Fig. 536).

(1) Locate engine control rigging pin mount (1) and attach with bolts (2 and 3) and washers (4).

(2) Lockwire all bolts using double twist method per 20-50-02.

NOTE: Parts of step (1), with parts shown in Fig. 548, belong to the engine controls linkage installation, reference 65-51534. This installation will be completed when the engine is fitted to the airplane.



REF.
FIG
1110

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
Q. (1)	1	MOUNT	65-51517-3		1
	1	MOUNT	65-51517-1 (OPT)		1
	2	BOLT	NAS1304-4HW		2
	3	BOLT	NAS1304-16HW		1
	4	WASHER	AN960PD416L		3
(2)		NO parts required			

OVERHAUL MANUAL

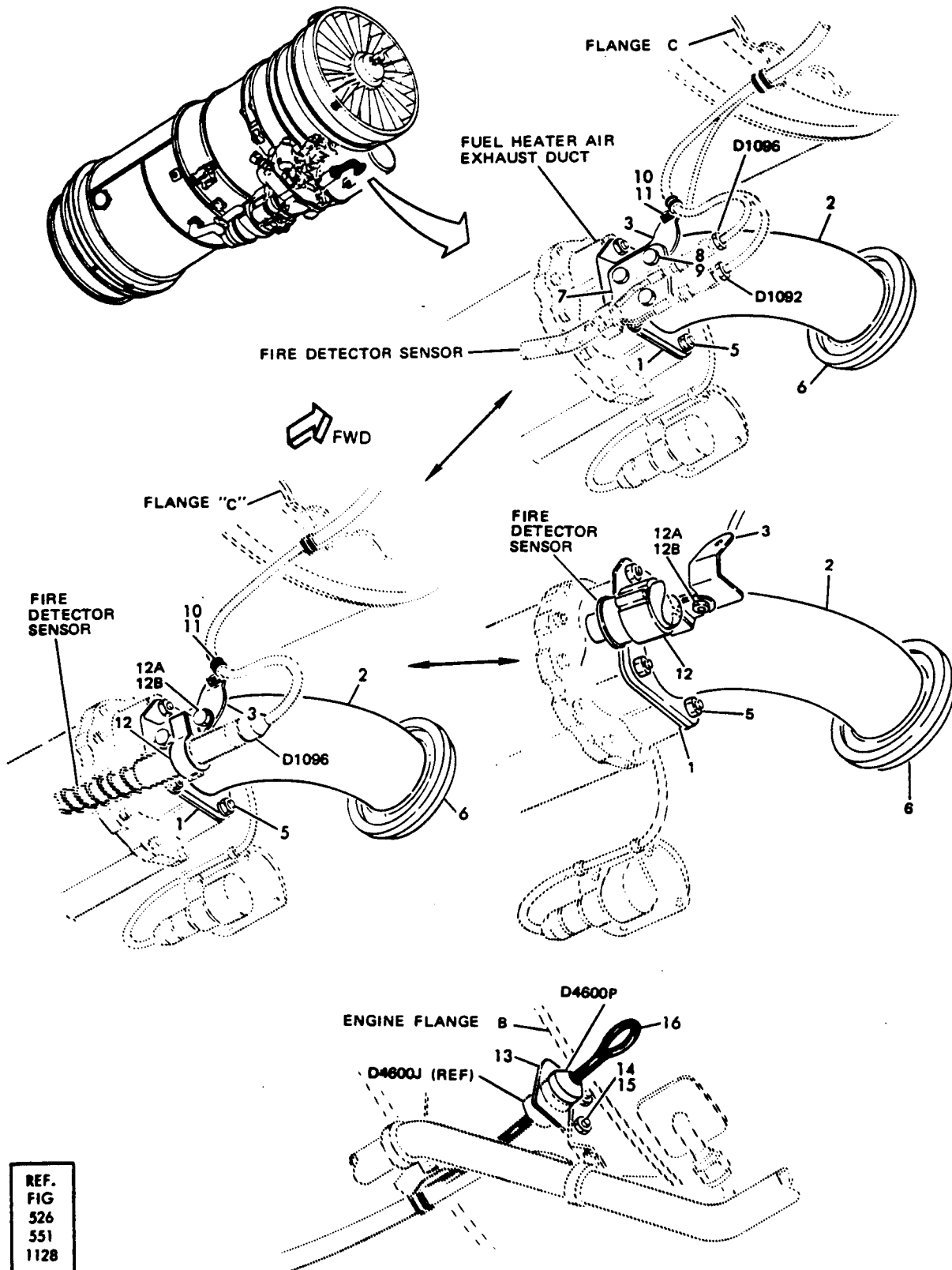
R. Install fuel heater air exhaust duct as follows: (Fig. 537)

- (1) Place gasket (1) over studs on fuel heater outlet pad.
- (2) Locate duct (2) and bracket (3). Attach with nuts (5). Place washer (4) beneath nut which is not common to bracket, and leave nut loose. This nut attaches bonding jumper when oil tank unit is installed.
- (3) Install seal (6) on end of duct (2).
- (4) 10-61096-115, -135, 10-62061-10 fire detector installation.
 - (a) Install bracket (7) with screws (8) and nuts (9).
 - (b) Attach end of sensor with screw (8) and nut (9).
 - (c) Make electrical connections to plugs D1092 and D1096 and lockwire connectors in place per wire bundle installation (Fig. 525).
 - (d) Install clamp (10) with screw (11).
- (5) 10-61096-35, -40, -105 fire detector installation.
 - (a) Install quick release clamp (12) with screws (12A) and nuts (12B).
 - (b) Install fire detector sensor in clamp (12).
 - (c) Connect electrical plug D1096 to fire detector plug per wire bundle installation (Fig. 525).
 - (d) Install clamp (10) with screw (11).
 - (e) Install bracket (13) on existing bracket on flange B with screws (14) and nuts (15).

NOTE: Bracket 69-48916-8 is used on power plants with hydraulically-operated thrust reversers; bracket 69-48916-6 is used on power plants with pneumatically-operated thrust reversers.

- (f) Connect plug D4600J of wire bundle W198 with plug D4600P of fire detector test receptacle jumper (16).

NOTE: Fire detector test receptacle (16) 61-30204-002 is used with hydraulically-operated thrust reversers.



REF.
 FIG
 526
 551
 1128

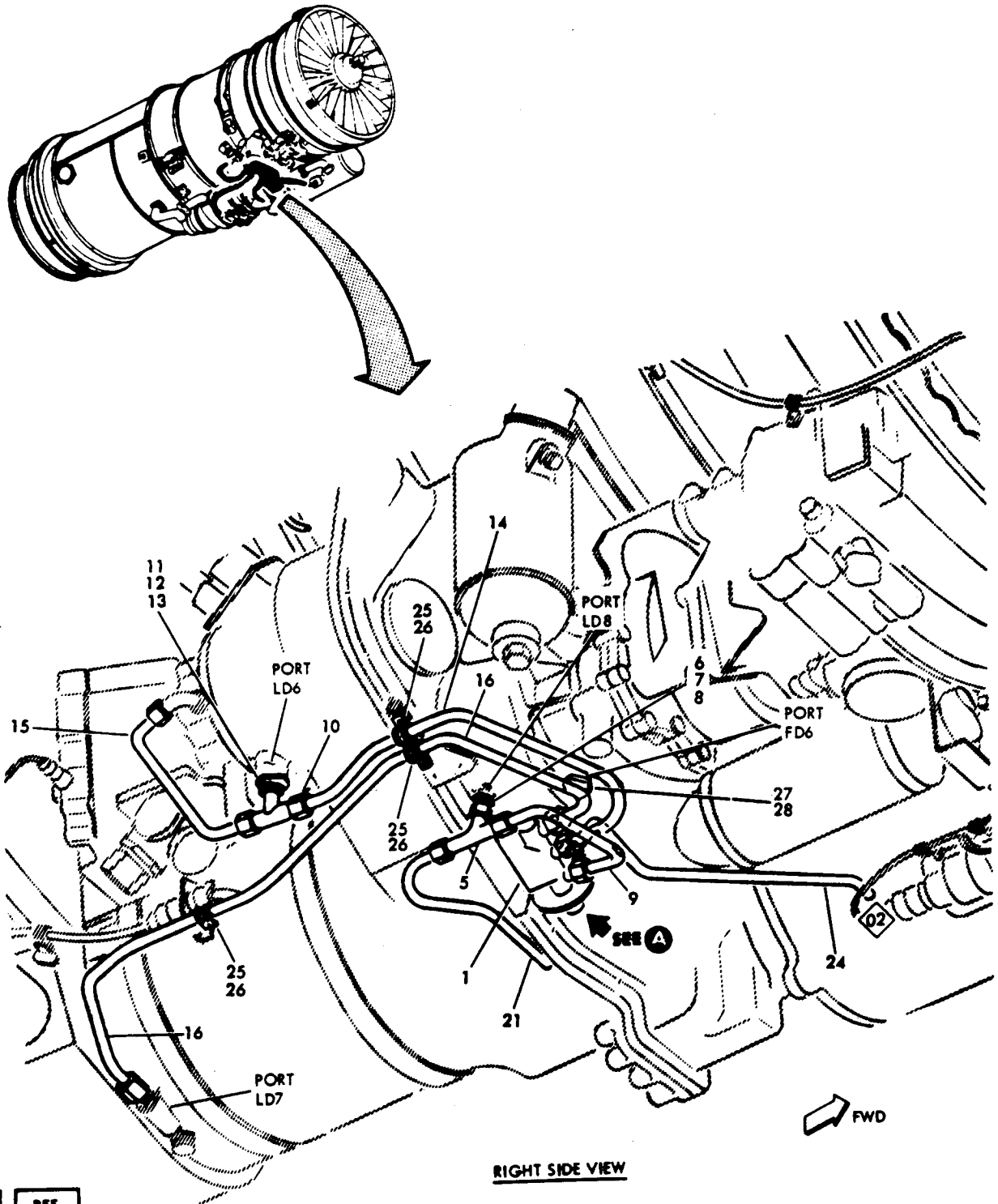
Fuel Heater Air Exhaust Duct Installation
 Figure 537 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
R.					
(1)	1	GASKET	66-15218-1		1
(2)	2	DUCT ASSY	69-36671-1		1
	3	BRACKET ASSY	69-52217-4		1
	3	BRACKET ASSY	69-52217-14		1
	4	WASHER	AN960C516L		1
	5	NUT	NAS679C5		4
	5	NUT	BACN10JC5C		4
(3)	6	SEAL	69-31608-1		1
(4)(a)	7	BRACKET	69-48918-7		1
	8	SCREW	BACS12CB3-7		3
	9	NUT	NAS679A3W		3
(b)		REFER TO PARTS STEP (a)			
(c)		NO PARTS REQUIRED			
(d)	10	CLAMP	BAC10BH5RW		1
	11	SCREW	BACS12CB3-6		1
(5)(a)	12	CLAMP ASSY	TA5000BH16HB		2
	12	CLAMP ASSY	TA5000BH14AT		1
	12A	SCREW	BAC12CB3-6		2
	12A	SCREW	NAS1801-3-6		2
	12B	NUT	NAS679A3W		2
	12B	NUT	BACN10JC3		2
(b)		NO PARTS REQUIRED			
(c)		NO PARTS REQUIRED			
(d)		REFER TO PARTS STEP (d)			
(e)	13	BRACKET	69-48918-8		1
	13	BRACKET	69-48918-6		1
	14	SCREW	BACS12CB3-6		2
	15	NUT	NAS679A3W		2
(f)	16	JUMPER ASSY, FIRE DETECTOR TEST RECEPTACLE	61-30204-001		1
	16	JUMPER ASSY, FIRE DETECTOR TEST RECEPTACLE	61-30204-002		1

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

S. Install engine drain plumbing as follows: (See figure 538.)

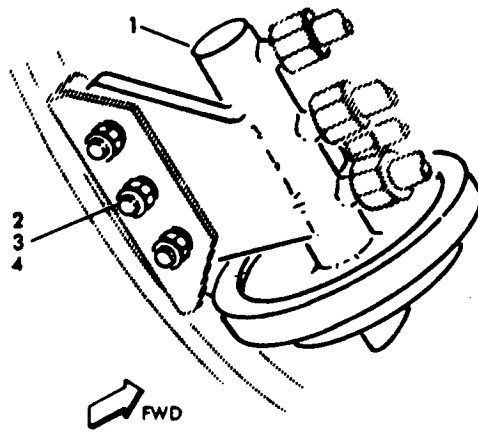
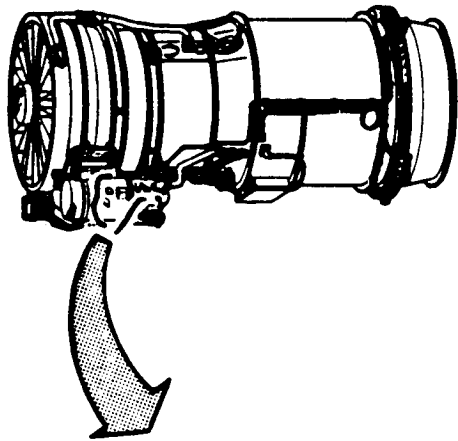
- (1) Install drain manifold (1) with screws (2), washers (3), and nuts (4) on existing bracket attached to forward side of gearbox.
- (2) Install nut (6) on tee (5). Install tee in fuel pump drain port LD8 with O-ring (7) and backup ring (8). Do not tighten nut.
- (3) Install tube assembly (9) between tee (5) and drain manifold (1). Tighten end fittings of tube assembly. Tighten nut (6).
- (4) Install nut (11) on tee (10) and install tee in accessory drive overboard drain port LD6 with packing (12) and backup ring (13). Do not tighten nut (11).
- (5) Install tube assembly (14) between tee (10) and drain manifold (1). Tighten end fittings of tube assembly. Tighten nut (11).
- (6) Install tube assembly (15) between fitting in hydraulic pump seal drain port and tee (10). Tighten end fittings of tube assembly.
- (7) Install tube assembly (16) between fitting in accessory drive drain port LD7 and drain manifold (1). Tighten end fittings of tube assembly.
- (8) Install nut (18) on elbow (17). Install elbow in gearbox overboard drain port LD5 with O-ring (19) and backup ring (20). Do not tighten nut (18).
- (9) Install tube assembly (21) between elbow (17) and aft port of tee (5). Tighten nut (18). Tighten end fittings of tube assembly. Install clamp (21A) with screw (21B).
- (10) Install union (22) and O-ring (23) in oil cup overflow drain port LD3. Tighten union.
- (11) Install tube assembly (24) between union (22) and drain manifold (1). Tighten end fittings of tube assembly.
- (12) Secure tube assemblies with clamps (25) and screws (26).
- (13) Install plug (27) and O-ring (28) in fuel pump seal drain port FD6.
- (14) Cover opening in drain manifold.



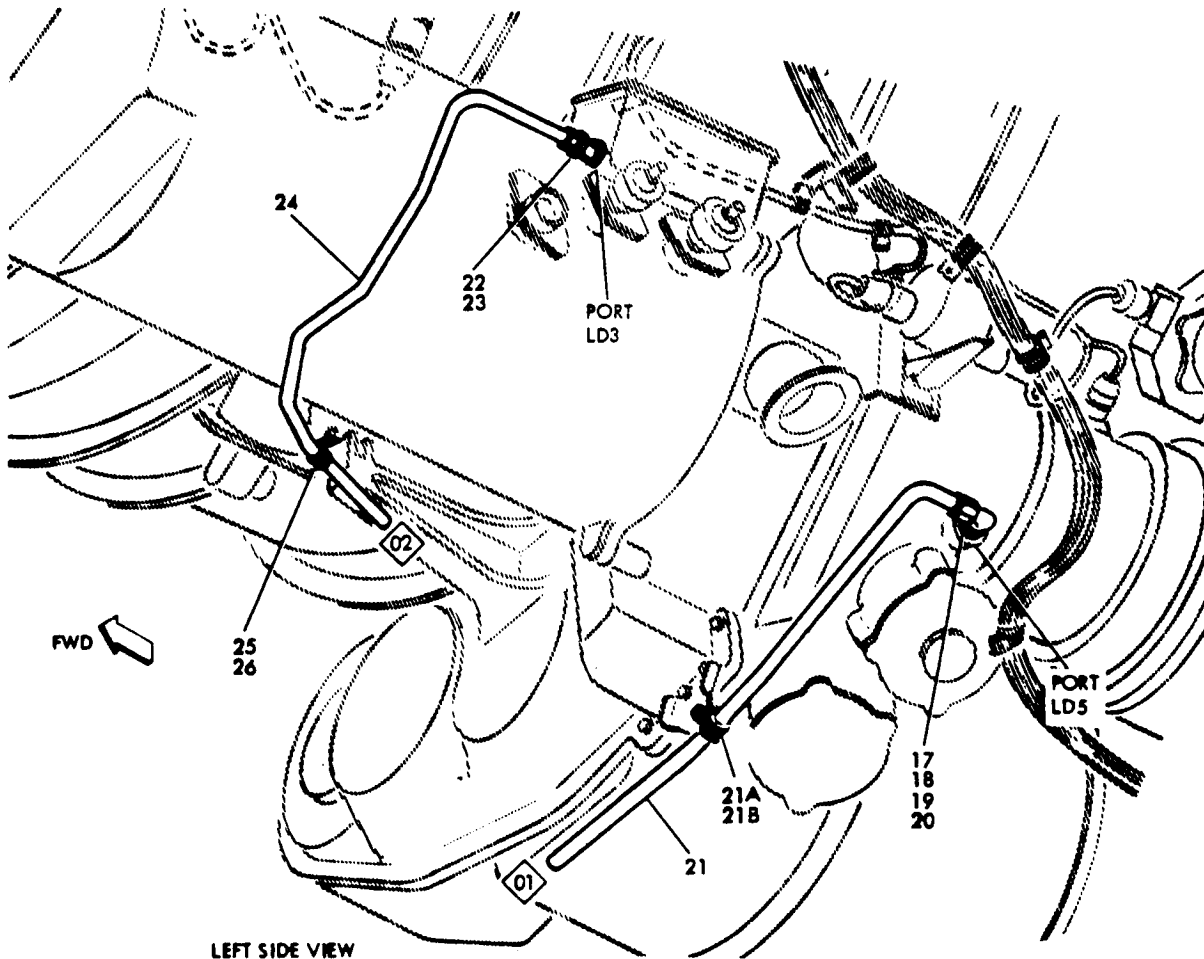
REF.
FIG
1119

Engine Drain Plumbing
Figure 538 (Sheet 1)

May 15/68



A



LEFT SIDE VIEW

Engine Drain Plumbing
 Figure 538 (Sheet 2)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
S.					
(1)	1	MANIFOLD, DRAIN	69-38973-1		1
	2	SCREW	BACS12CB3-6		3
	3	WASHER	AN960-10L		3
	4	NUT	NAS679A3W		3
(2)	5	TEE	118-10555		1
	6	NUT	AN6289-6		1
	7	O-RING	BACP11P-B6		1
	8	RING, BACKUP	MS28777-6		1
(3)	9	TUBE ASSEMBLY	65-55475-5		1
(4)	10	TEE	MS21909-6		1
	11	NUT	AN6289-6		1
	12	PACKING	NAS1612-6		1
	13	RING, BACKUP	MS28777-6		1
(5)	14	TUBE ASSEMBLY	65-55475-24		1
(6)	15	TUBE ASSEMBLY	65-55475-8		1
(7)	16	TUBE ASSEMBLY	65-55475-23		1
(8)	17	ELBOW	MS21926-6		1
	18	NUT	AN6289-6		1
	19	O-RING	BACP11P-B6		1
	20	RING, BACKUP	MS28777-6		1
(9)	21	TUBE ASSEMBLY	65-55475-25		1
	21A	CLAMP	BACC10HSO6		1
	21B	SCREW	BACS12CB3-7		1
(10)	22	UNION	MS21902-6		1
	23	O-RING	BACP11P-B6		1
(11)	24	TUBE ASSEMBLY	65-55475-4		1
(12)	25	CLAMP	BACC10HSO6		4
	26	SCREW	BACS12CB3-7		4
(13)	27	PLUG	AN814-6CL		1
	28	O-RING	BACP11P-B6		1
(14)		NO PARTS REQUIRED			

T. Install fuel flowmeter as follows: (Fig. 539)

- (1) Remove aft support bracket (1) installed by engine manufacturer.

NOTE: Place a waste container beneath flowmeter to catch fluid which drains when flowmeter adapter is detached.

- (2) Remove four bolts which attach lockplate (2) on forward end of flowmeter adapter installed by engine manufacturer.
- (3) Detach tee connector (3), installed by engine manufacturer, from fuel supply tube, and remove entire flowmeter assembly.
- (4) Install bracket assembly (4) as follows:
 - (a) Remove existing flange bolts.
 - (b) Locate bracket and reinstall flange bolts.
 - (c) Tighten flange bolt within torque range of 110 to 150 pound-inches and reinstall lockwire on bolts.
- (5) Attach flowmeter wire bundle to bracket assembly (4) with clamp (5) and screw (6).
- (6) Remove lockplate (2) and connector (7) from front end of flowmeter adapter.
- (7) Remove tee connector (3) from aft end of flowmeter adapter.

CAUTION: DO NOT ALLOW TORQUE TO BE TRANSMITTED THROUGH FLOWMETER BODY WHILE TIGHTENING END FITTING (7).

- (8) Install new O-ring (8), O-ring (9), and backup ring (10), on connector (7). Apply Dow Corning Molykote, Type G-N, to threads of fitting (7). Install fitting (7), lockplate (2). Tighten to 3200-3500 pound-inches (P&W fitting 447884) or to 570-630 pound-inches (P&W fitting 773086) while gripping wrenching flats on meter body adjacent to fitting port. Lockwire fitting (7).

- (9) If 10-60507-14 or -17 flowmeter is used, perform steps (a) and (b). If 10-60507-1 or -6 flowmeter is used, perform step (c) only.
- (a) Install O-ring (13) on adapter (12), and install adapter in aft end of flowmeter (11).
- (b) Install O-ring (14) and backup ring (15) on tee connector (3), and install tee connector in adapter (12). Tighten jamnut (16) on tee connector, and install lockwire MS20995N32 or MS20995C32 on jamnut (double twist method).
- (c) Install O-ring (14) and backup ring (15) on tee connector (3) and install in aft end of flowmeter (11). Tighten jamnut (16) on tee connector, and install lockwire on jamnut.

- (10) Attach bonding jumper (17) to flowmeter attaching plate with screw (18), washer (19), and nut (20). Tighten nut (20) to 20-30 lb-in.

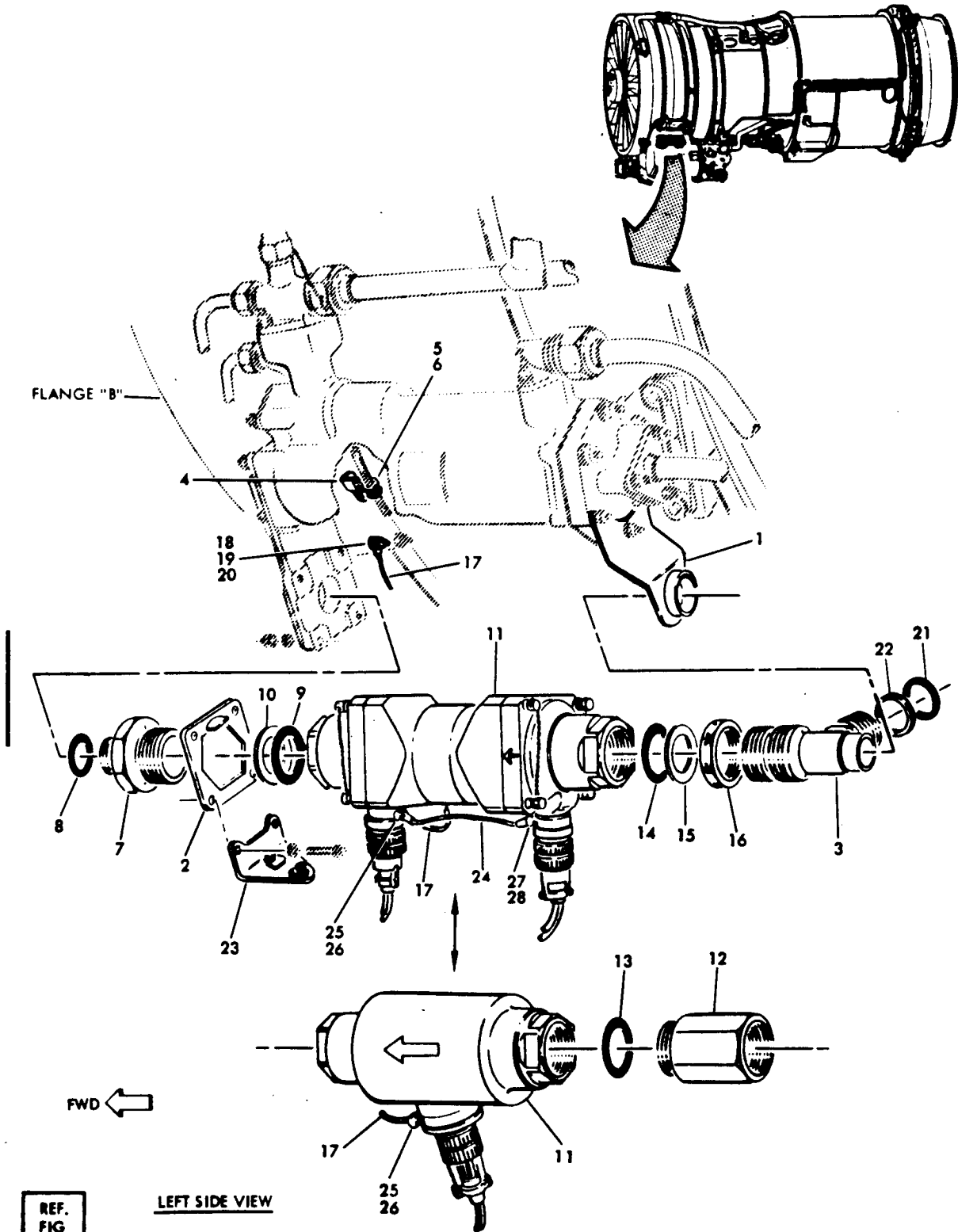
NOTE: Maximum allowable resistance across electrical bond is 0.0025 ohms (2.5 milliohms).

- (11) Install retainer (22) and silastic (21) on tee connector and locate flowmeter on engine with direction of flow indicator forward. Attach tee connector to fuel inlet tube.
- (12) Locate bracket assembly (23). Attach bracket assembly (23) and lockplate (2) by reinstalling attaching hardware removed in step (1).
- (13) Reinstall aft support bracket (1), removed in step (2).
- (14) Attach electrical connectors and lockwire in place.

NOTE: If 10-60507-14 or -17 flowmeter is used, only one connector is used.

- (15) If 10-60507-1, or 10-60507-6 flowmeter is used, perform steps (a) and (b) only. If 10-60507-14 or -17 flowmeter is used, perform step (c) only.
- (a) Attach bonding jumpers (17 and 24) to forward bonding lug on flowmeter with screw (25) and nut (26). Tighten nut (26) to 20-30 lb-in.
- (b) Attach free end of bonding jumper (24) to aft bonding lug on flowmeter with screw (27) and nut (28). Tighten nut (28) to 20-30 lb-in.
- (c) Attach bonding jumper (17) to bonding lug on flowmeter with screw (25) and nut (26).

NOTE: Maximum allowable resistance across electrical bond is 0.0025 ohm (2.5 milliohms).



Fuel Flowmeter Installation
 Figure 539 (Sheet 1)

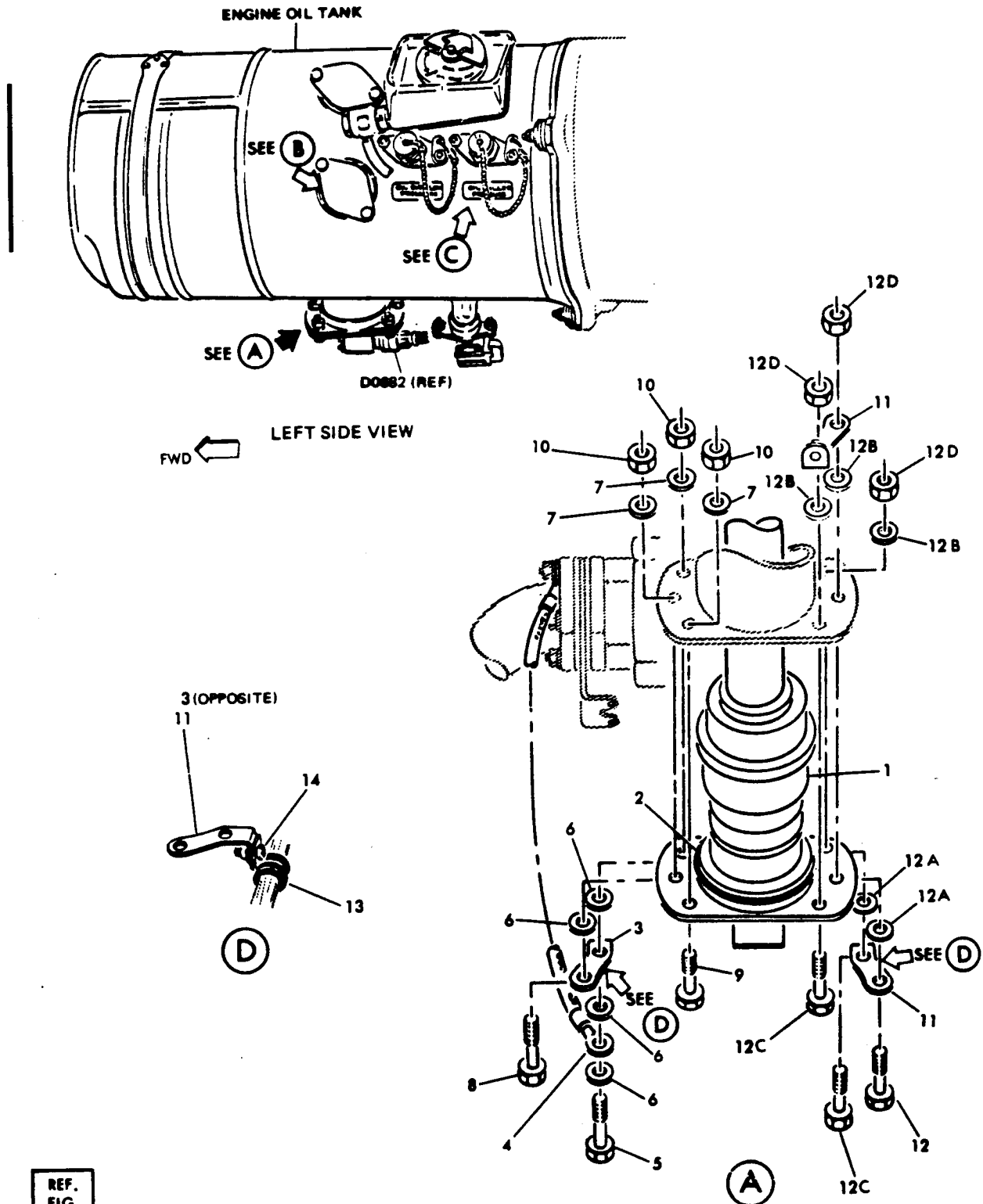
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
T.					
(1)	1	BRACKET	P&W INSTALLED		RF
(2)	2	PLATE, LOCK	P&W INSTALLED		RF
(3)	3	CONNECTOR, TEE	P&W INSTALLED		RF
(4)	4	BRACKET ASSEMBLY	65-22486-32		1
(5)	5	CLAMP	BACC10BH5RW		1
	6	SCREW	BACSL2CB3-7		1
(6)	7	CONNECTOR	P&W INSTALLED		RF
(7)		NO PARTS REQ			
(8)	8	O-RING	MS9021-021		1
	9	O-RING	MS9020-20		1
	10	RING, BACKUP	MS9058-20		1
(9)(a)	11	FLOWMETER	10-60507-1		1
	11	FLOWMETER	10-60507-6		1
	11	FLOWMETER	10-60507-14		1
	11	FLOWMETER	10-60507-17		1
	12	ADAPTER	69-35244-1 (USED ONLY WITH 10-60507-14 AND -17)		1
	13	O-RING	MS9020-20 (USE ONLY WITH 69-35244-1)		1
(b)	14	O-RING	MS9020-20		1
	15	RING, BACKUP	MS9058-20		1
	16	JAMNUT	P&W INSTALLED		RF
(c)		NO PART REG			
(10)	17	JUMPER, BONDING	BACJ40A20-9		1
	18	SCREW	BACSL2CB3-9		1
	19	WASHER	AN960C10L		1
	20	NUT	NAS679A3W		1

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(11)	21	SILASTIC	226195		1
	22	RETAINER	354117		1
(12)	23	BRACKET ASSEMBLY	66-15227-5		1
(13)		No parts required			
(14)		No parts required			
(15)	24	JUMPER, Bonding	BACJ40A20-5 (Used with 10-60507-1, or -6 only)		1
	25	SCREW	BACS12CB3-7		1
	26	NUT	NAS679A3W		1
	27	SCREW	BACS12CB3-6 (Used with BACJ40A20-5 only)		1
	28	NUT	NAS679A3W (Used with BACJ40A20-5 only)		1

OVERHAUL MANUAL

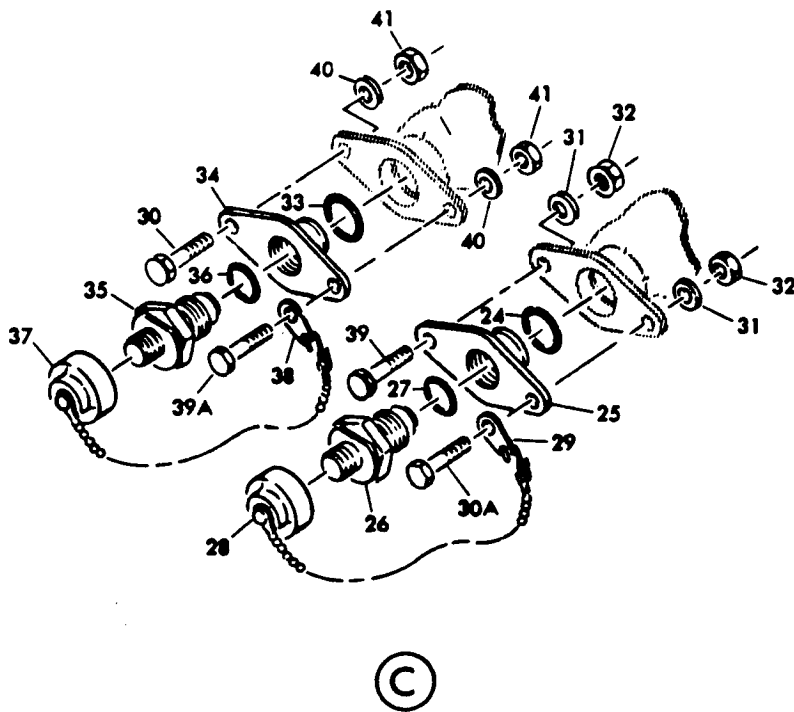
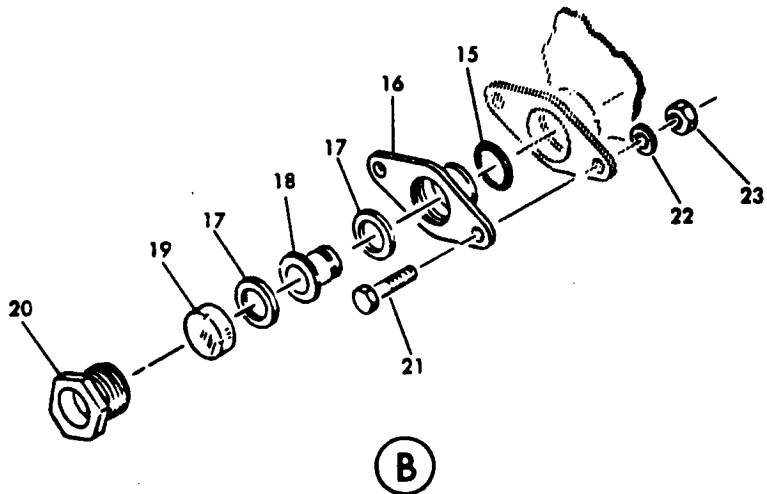
- U. Install oil tank accessory components on engine oil tank as follows:
(Fig. 540)
- (1) Place O-ring (2) in groove in tank unit (1). Insert telescoped end of unit through port in bottom of oil tank (detail A).
 - (2) Locate bracket (3) and bonding jumper (4). Locate two washers (6) beneath bracket at bolt holes, and install bolts (5, 8, and 9), with two washers (6) and three washers (7), and three nuts (10). Attach free end of bonding jumper (4) to fuel heater exhaust flange with existing nut (installed in par. R.). Tighten nuts (10) to 65-70 lb-in. Maximum allowable resistance is 0.001 ohms (1 milliohm).
 - (3) Locate brackets (11) and two washers (12A) as shown in detail A. Install bolt (12), two bolts (12C), three washers (12B), and three nuts (12D).
 - (4) Attach electrical plug D0682 to connector on tank unit (1). Install clamps (13) and screws (14).
 - (5) Install oil quantity sight glass unit (two places) as follows (detail B).
 - (a) Install O-ring (15) in groove on sight gage fitting (16).
 - (b) Place washer (17) on sight gage reflector (18), and insert reflector in sight gage fitting (16).
 - (c) Place glass (19) and washer (17) in cap (20), and screw cap into sight gage fitting (16). Tighten cap (20) to 50 lb-in.
 - (d) Install units assembled in steps (a) thru (c) in forward bosses on outer surface of tank with bolts (21), washers (22), and nuts (23).
 - (e) Lockwire items (20) and (21) together per 20-50-02.
 - (6) Install pressure filling fitting assemblies as follows (detail C):
 - (a) Place O-ring (24) in groove on fitting (25).
 - (b) Install bulkhead disconnect coupling half (26) and O-ring (27) in fitting (25).
 - (c) Install dust cap (28) on coupling half (26).

- (d) Install unit assembled in steps (a) through (c) in aft boss on upper outside surface of tank.
- (e) Locate clip (29) and install bolts (30 and 30A), washers (31), and nuts (32). Attach chain from dust cap (28) to clip (29).
- (f) Place O-ring (33) in groove on fitting (34).
- (g) Install quick-disconnect coupling half (35) and O-ring (36) in fitting (34).
- (h) Install dust cap (37) on quick-disconnect coupling.
- (i) Install unit assembled in steps (f) through (h) in forward boss on upper outside surface of oil tank.
- (j) Locate clip (38) and install bolts (39 and 39A), washers (40), and nuts (41). Attach chain from dust cap (37) to clip (38).
- (k) Install Metal-Cal (42) beneath forward pressure port as directed in Subject 20-50-05.
- (l) Install Metal-Cal (43) beneath aft pressure port as directed in Subject 20-50-05.



REF.
FIG.
1107

Oil Tank Accessory Components Installation
Figure 540 (Sheet 1)



Oil Tank Accessory Components Installation
Figure 540 (Sheet 2)

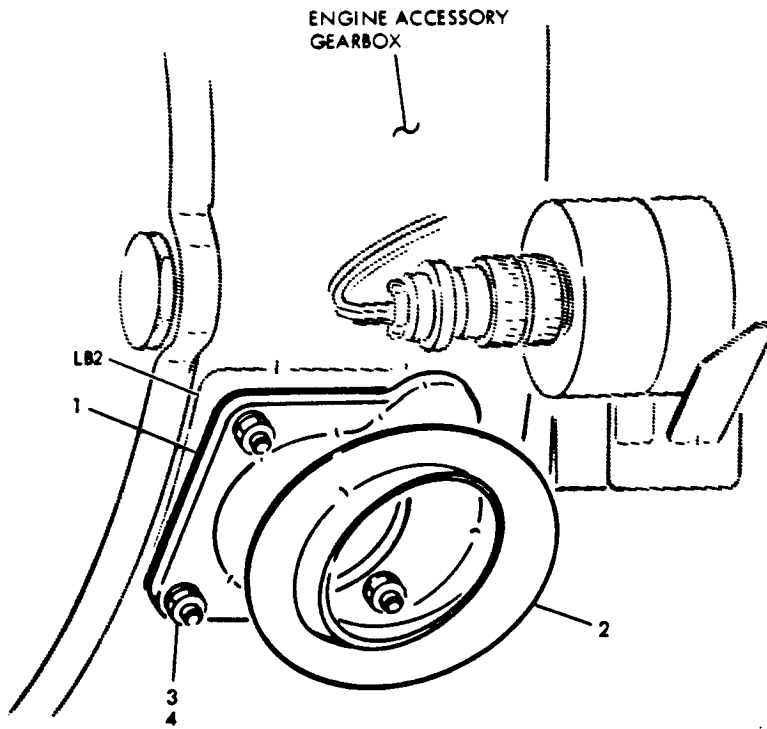
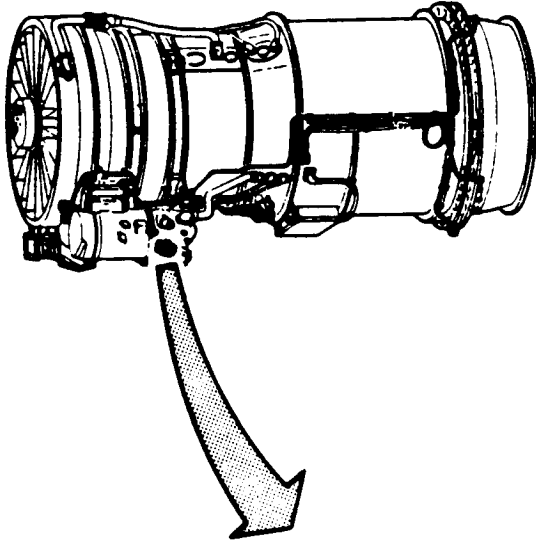
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
U. (1)	1	TANK UNIT, Oil Quantity Indication System	10-60722-4		1	
	2	O-RING	BACP11P228		1	
	(2)	3	BRACKET	69-45804-3		1
		4	JUMPER, Bonding	BACJ40K7A6A6		1
		5	BOLT	NAS1104-8		1
		6	WASHER	AN960C416L		4
		7	WASHER	AN960C416		3
		8	BOLT	NAS1104-6		3
		9	BOLT	NAS1104-4		1
		10	NUT	NAS679C4W		3
(3)	11	BRACKET	69-45804-2		2	
	12	BOLT	NAS1104-7		1	
	12A	WASHER	AN960C416L		2	
	12B	WASHER	AN960C416		3	
	12C	BOLT	NAS1104-6		2	
	12D	NUT	NAS679C4W		3	
(4)	13	CLAMP	BACC10BH4RW		2	
	14	SCREW	SCREW		2	
(5)(a)	15	O-RING	BACP11P26		2	
	16	FITTING, Sight Gage	66-15330-1		2	
(b)	17	WASHER	BACW10H101		4	
	18	REFLECTOR	3-58452		2	
(c)	19	GLASS	BACG21A10		2	
	20	CAP	BACC14B1CH		2	
(d)	21	BOLT	NAS1304-2HW		4	
	22	WASHER	AN960C416L		4	
	23	NUT	NAS679C4W		4	
(e)		No parts required				
(6)(a)	24	O-RING	BACP11P21		1	
	25	FITTING, Pressure Filling	69-31661-1		1	
	25	FITTING, Pressure Filling	66-16394-1 (Optional)		1	

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(b)	26	COUPLING HALF, Bulkhead Disconnect	OMP2506-2		1
	27	O-RING	BACP11P-B6		1
(c)	28	CAP, Dust	OMP2506-4		1
(d)		No parts required			
(e)	29	CLIP	66-16435-1		1
	30	BOLT	NAS1104-3		1
	30A	BOLT	NAS1104-4		1
	31	WASHER	AN960C416L		2
	32	NUT	NAS679C4W		2
(f)	33	O-RING	BACP11P21		1
	34	FITTING	69-31661-2		1
	34	FITTING	66-16394-2		1
(g)	35	COUPLING HALF, Quick-Disconnect	OMP2505-2		1
	36	O-RING	BACP11P-B8		1
(h)	37	CAP, Dust	OMP2505-4		1
(i)		No parts required			
(j)	38	CLIP	66-16435-1		1
	39	BOLT	NAS1104-3		1
	39A	BOLT	NAS1104-4		1
	40	WASHER	AN960C416L		2
	41	NUT	NAS679C4W		2
(k)	42	METAL-CAL	BACM10P6GT		1
(l)	43	METAL-CAL	BACM10P6GS		1

- V. Install gearbox overboard vent fitting as follows: (See figure 541.)
- (1) Install gasket (1) over studs on gearbox overboard breather pad LB2.
 - (2) Install fitting (2) and secure with washers (3) and nuts (4).
 - (3) Cover opening in fitting.



LEFT SIDE VIEW

REF.
FIG
1137

Gearbox Overboard Vent Fitting Installation
Figure 541 (Sheet 1)

737
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

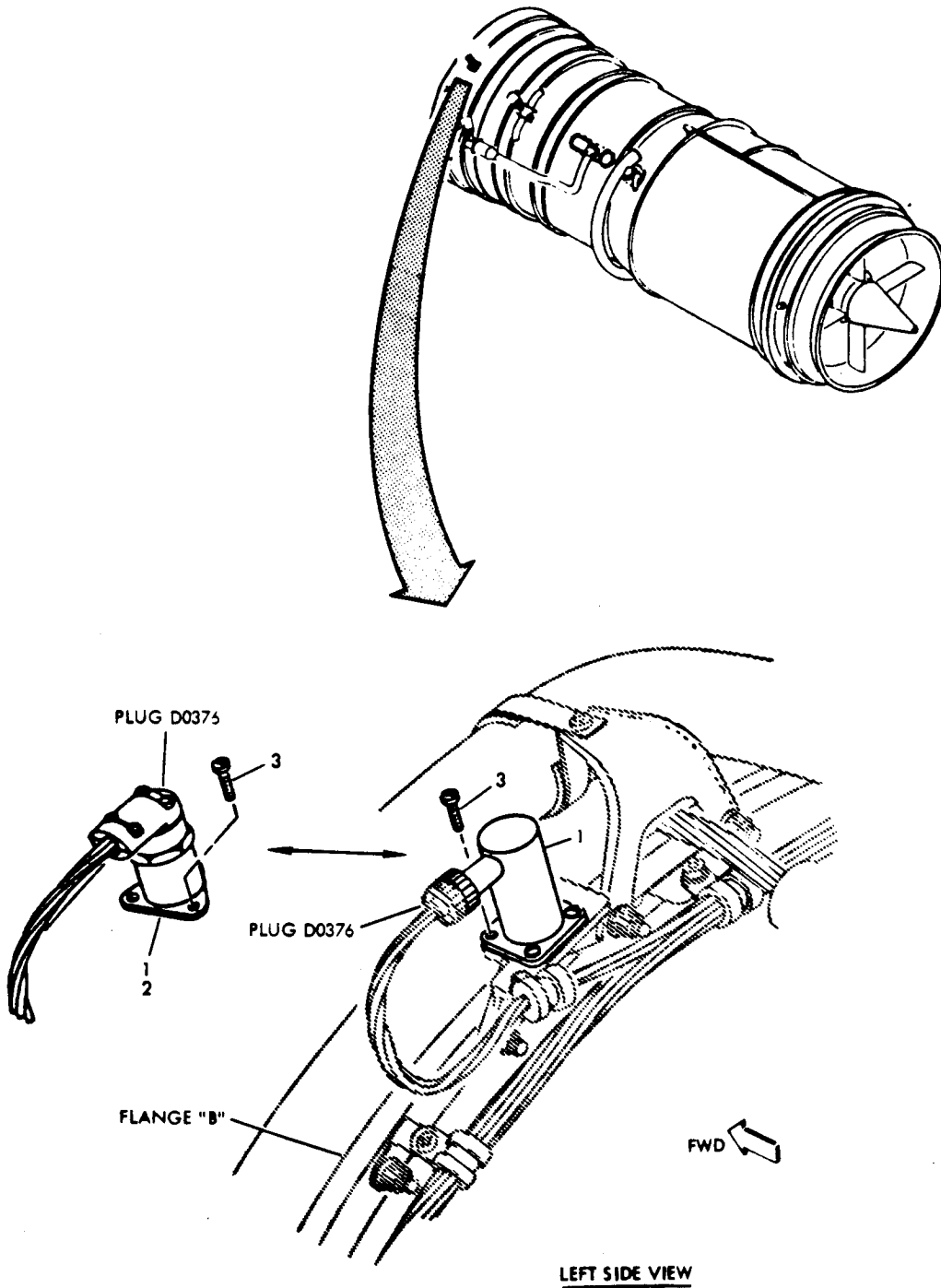
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
V. (1)	1	GASKET	66-14330-1		1
(2)	2	FITTING	65-61732-1		1
	2	FITTING	65-63375-1 (Optional)		1
	3	WASHER	AN960D416		4
	4	NUT	NAS679A4W		4
(3)		No parts required			

W. Install inlet vibration pickup as follows (Fig. 542):

NOTE: Perform step (1) only if vibration pickup P/N 1784340-10 is used.

- (1) Attach vibration pickup (1) to adapter (2). Secure pickup to adapter with lockwire.
- (2) Locate vibration pickup on existing bracket on flange B to left of engine centerline. Install screws (3).
- (3) Lockwire all screws.
- (4) Attach electrical plug D0376 to connector on pickup and lockwire in place.

NOTE: Use of inlet vibration pickup on SV airplanes is limited.
If not used, cap and stow D0376.



REF.
FIG
1106

Inlet Vibration Pickup Installation
Figure 542 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
W. (1)	1	PICKUP, VIBRATION	4-125-0001		1	
	1	PICKUP, VIBRATION	4-126-0001		1	
	1	PICKUP, VIBRATION	WL-800		1	
	1	PICKUP, VIBRATION	MODEL 750		1	
	1	PICKUP, VIBRATION	1784340-10		1	
	1	PICKUP, VIBRATION	MODEL 9145		1	
	1	PICKUP, VIBRATION	WL-750		1	
	1	PICKUP, VIBRATION	1784340-60		1	
	1	PICKUP, VIBRATION	144-904-000-01		1	
	1	PICKUP, VIBRATION	144-904-000-04		1	
	1	PICKUP, VIBRATION	144-904-000-05		1	
	2	ADAPTER	1510376-2 (USE WITH 1784340-10 ONLY)		1	
	(2)	3	SCREW	AN500AD6-8		4
		3	SCREW	AN500AD8-7 (USE WITH 1510376-2 ONLY)		3
(3)		NO PARTS REQUIRED				
(4)		NO PARTS REQUIRED				

X. Install engine oil pressure transmitter and switch as follows:
(Fig. 543)

(1) For 65-50882-1, -2 engine oil pressure and low pressure warning transmitter and switch installations:

(a) On installations vented to engine gearbox:

1) Install jamnut (1) on elbow (2), and install elbow in gearbox transmitter oil vent port LV3 with O-ring (3) and backup ring (4). Do not tighten jamnut.

(b) On installations vented to ambient pressure:

1) Install plug (4A) with O-ring (4B) in gearbox transmitter oil vent port LV3.

(c) Install restrictor union (5) in oil pressure line low pressure warning port LP3 with O-ring (6). Tighten union.

(d) Locate transmitter (7) and switch assembly (7A) on upper left side of flange "D".

(e) Attach transmitter (7) to existing bracket on flange "D" with screws (8), washers (9), and nuts (10).

NOTE: Make electrical bond between parts. Maximum allowable resistance between transmitter or switch and basic structure is 0.0025 ohms (2.5 milliohms).

(f) Attach pressure switch (7A) to existing bracket on flange "D" with two screws (11), washers (12), and nuts (13).

NOTE: Make electrical bond between parts. Maximum allowable resistance between transmitter or switch and basic structure is 0.0025 ohms (2.5 milliohms).

(g) Attach free end of hose from elbow in pressure port of switch to union (5).

(h) Align all hoses in transmitter and switch assembly. Adjust position of transmitter and tighten T-bolt in shockmount. Tighten fittings in ports of switch and secure with lockwire.

(i) Tighten all hose end fittings in installation.

(j) On installations vented to engine gearbox:

1) Tighten jamnut (1).

2) Install tube assembly (14) between elbow (2) and fitting in VENT port of switch. Tighten end fittings of tube assembly.

OVERHAUL MANUAL

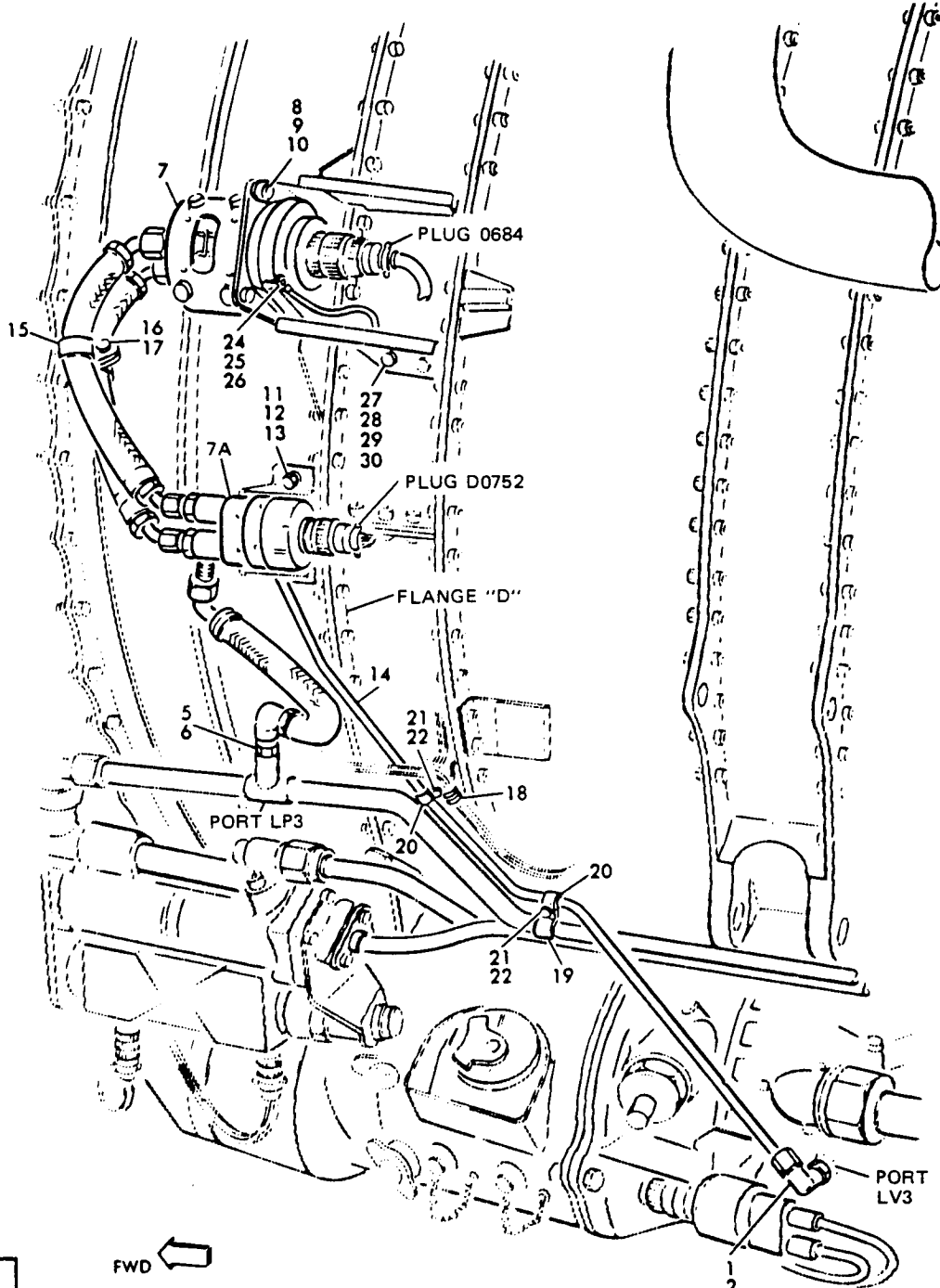
- 3) Clamp pressure hose and vent hose between transmitter and switch with floating clamps (15, 15A), screw (16) and nut (17).
- 4) Attach tube assembly (14) to existing wire bundle and tube with clamps (18, 19, 20), screws (21), and nuts (22). Tighten nuts (22) at wire bundle to 25-35 lb-in.
- (k) Attach plug D0752 to electrical connector on switch.
- (l) Attach plug D0684 to electrical connector on oil pressure transmitter.
- (m) Remove screw from pressure transmitter case. Attach end of bonding jumper (23) to transmitter case with screw (24), washer (25), and lockwasher (26).
- (n) Attach free end of bonding jumper to transmitter mount bracket with screw (27), washer (28), washer (29), and nut (30). Tighten nut (30) to 20-30 lb-in.

NOTE: Maximum allowable resistance between jumper (23) and structure is 0.0025 ohms (2.5 milliohms).

- (2) 65-50882-3 engine oil pressure and low pressure warning transmitter and switch installation:
 - (a) Locate and attach transmitters (31) to existing brackets on upper left side of flange "D" and flange "G" using nut (32).
 - (b) Install reducer (33) and vent port (34) with O-rings (35) on transmitter located on flange "D".
 - (c) Locate and attach pressure switch (36) to existing bracket on upper left side of flange "D" using screws (37), washers (38), and nuts (39).
 - (d) Install restrictor union (40) in oil pressure line low pressure warning port LP3 with O-ring (41). Tighten union.
 - (e) Attach free end of hose from elbow in pressure port of switch to union (40).
 - (f) Attach tube assembly (42) to pressure port of transmitter located on flange "D" and to OIL port of switch.
 - (g) Install reducers (43) with O-rings (44) in pressure ports of transmitter located on flange "G".
 - (h) Attach tube assemblies (45, 46) to reducers (43).
 - (i) Attach tube assemblies (45, 46) to engine oil differential pressure switch (Fig. 545).

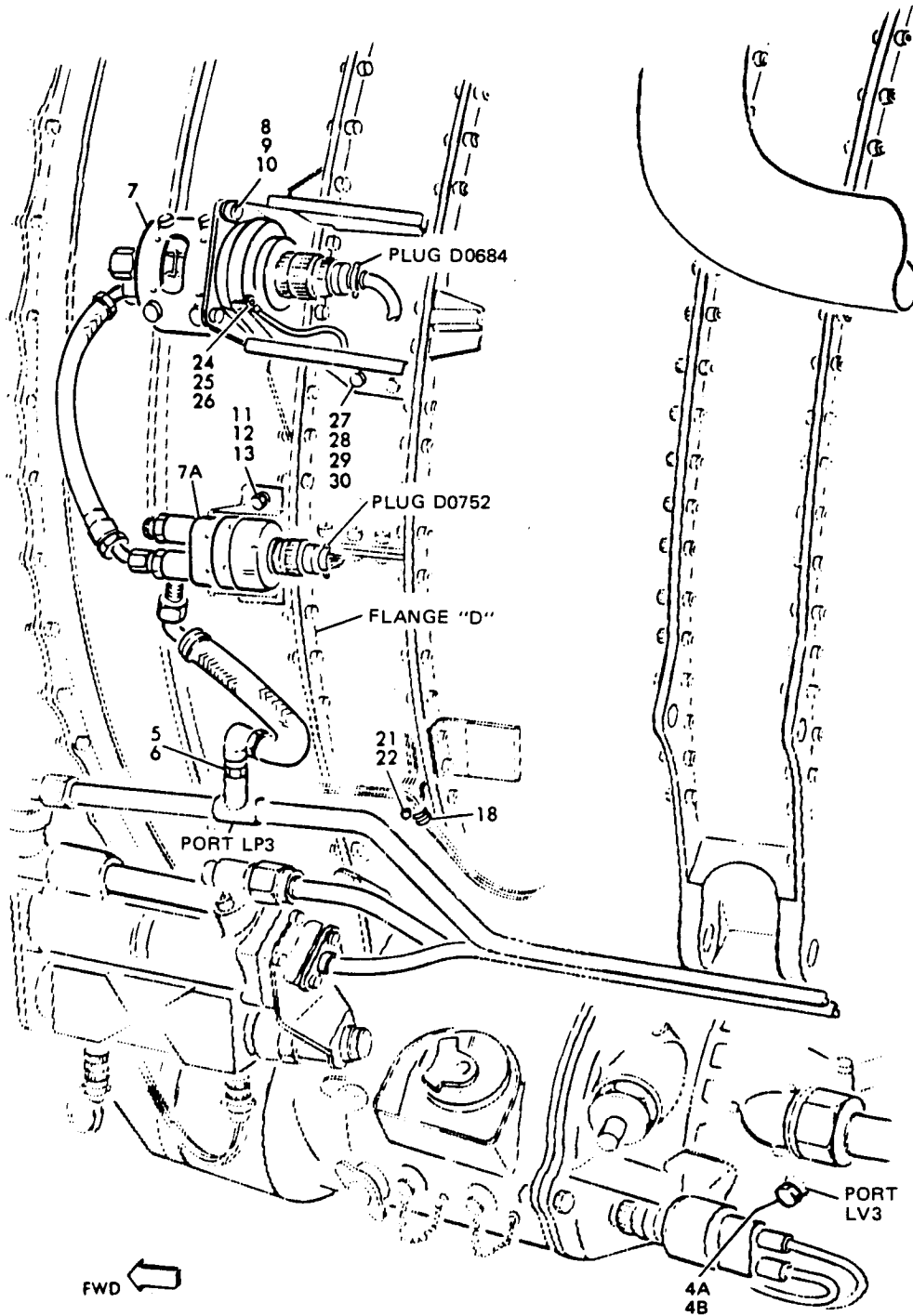
OVERHAUL MANUAL

- (j) Install floating clamps (47) on tube assemblies (45, 46) between transmitter and differential pressure switch using screws (48) and nuts (49).
- (k) Tighten all end fittings.



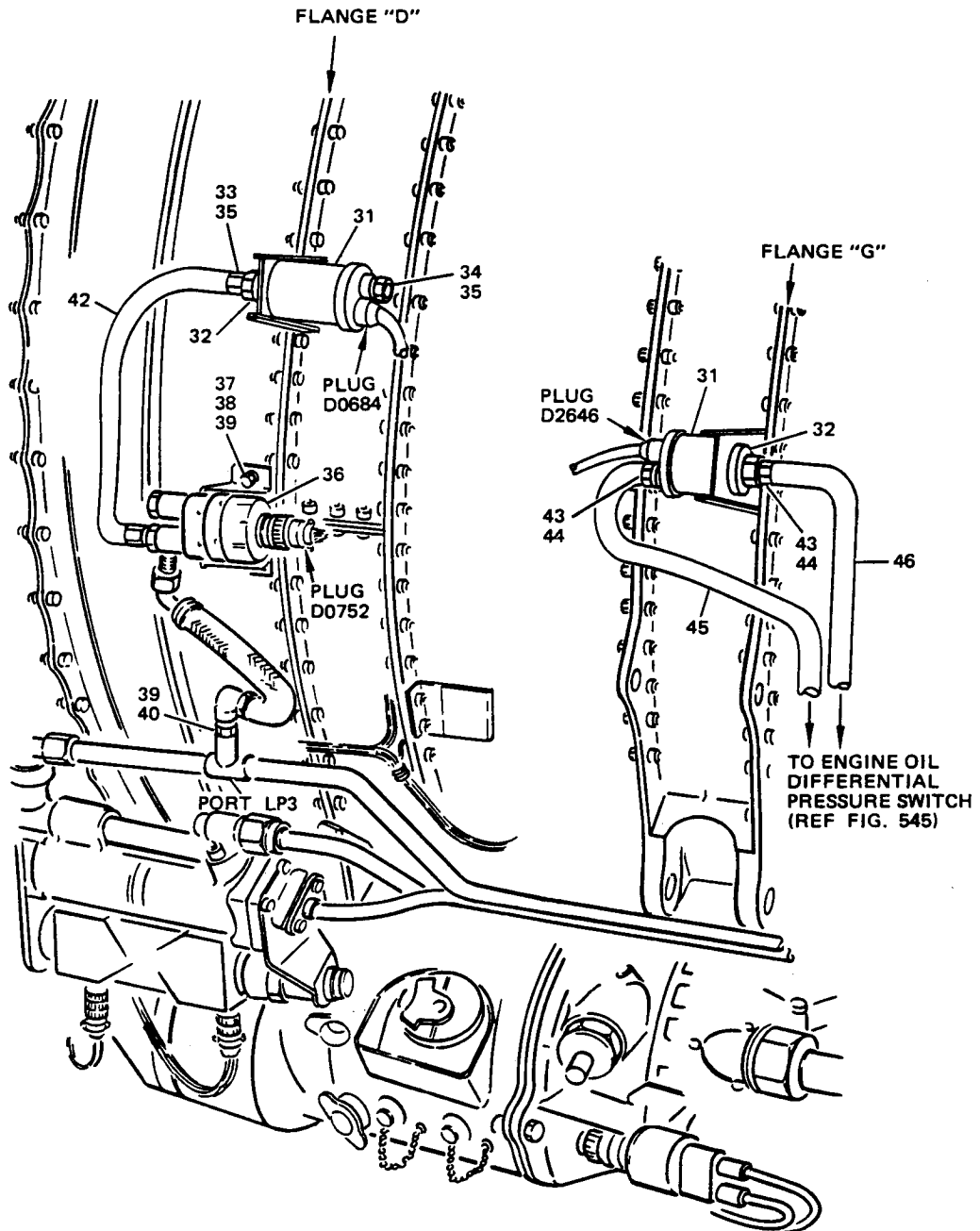
REF.
FIG
505
1115
1126

65-50882-1, -2
GEARBOX VENTING CONFIGURATION
LEFT SIDE VIEW



65-50882-1, -2
AMBIENT VENTING CONFIGURATION
LEFT SIDE VIEW

REF.
FIG
505
1115
1126



65-50882-3 INSTALLATION

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
X.					
(1)(a)	1	JAMNUT	AN6289-4		1
	2	ELBOW	MS21908-4		1
	3	O-RING	BACP11PB4		1
	4	RING, BACKUP	MS28777-4		1
(b)	4A	PLUG	MS9015-04		1
	4B	O-RING	MS9387-04		1
(c)	5	UNION, RESTRICTOR	D9798		1
	5	UNION, RESTRICTOR	1-01288		1
	5	UNION, RESTRICTOR	69-64545-1		1
	5	UNION, RESTRICTOR	69-64545-4		1
	5	UNION, RESTRICTOR	1-01188		1
	5	UNION, RESTRICTOR	118-10398		1
	6	O-RING	BACP11PB6		1
(d)	7	TRANSMITTER, OIL PRESSURE	ST104Z *[1]		1
	7	TRANSMITTER, OIL PRESSURE	ST104AB *[1]		1
	7A	SWITCH, PRESSURE	42D107A1		1
(e)	8	SCREW	BACS12CB3-9		4
	9	WASHER	AN960-10L		4
	10	NUT	BACN10JC3		4
(f)	11	SCREW	BACS12CB3-7		2
	12	WASHER	AN960-10L		2
	13	NUT	BACN10JC3		2
(g)		NO PARTS REQUIRED			
(h)		NO PARTS REQUIRED			
(i)		NO PARTS REQUIRED			
(j)1)		NO PARTS REQUIRED			
2)	14	TUBE ASSEMBLY	65-58135-14		1
3)	15	CLAMP	BACC10HS10		1
	15A	CLAMP	BACC10HS12		1
	16	SCREW	BACS12CB3-7		1
	17	NUT	BACN10JC3		1

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
X.					
	4)	18 CLAMP	BACC10HS10		1
		19 CLAMP	BACC10HS12		1
		20 CLAMP	BACC10HSO4		2
		21 SCREW	BACS12CB3-7		2
		22 NUT	BACN10JC3		2
	(k)	NO PARTS REQUIRED			
	(l)	NO PARTS REQUIRED			
	(m)	23 JUMPER, BONDING	MS25083-2AB5		1
		24 SCREW	BACS12CB06-4		1
		25 WASHER	AN960-6L		1
		26 LOCKWASHER	MS35338-41		1
	(n)	27 SCREW	BACS12CB3-5		1
		28 WASHER	AN960-10L		1
		29 LOCKWASHER	MS35338-43		1
		30 NUT	BACN10JC3		1
	(2)(a)	31 TRANSMITTER, OIL PRESSURE	418-20044		2
		32 NUT	BACN10DR12J		2
	(b)	33 REDUCER	MS21916J6-4		1
		34 VENT PORT	66-13955-875		1
		35 O-RING	BACP11PB4		2
	(c)	36 SWITCH, PRESSURE	42D107A1 *[1]		1
		37 SCREW	NAS1801-3-7		2
		38 WASHER	AN960-10L		2
		39 NUT	BACN10JC3		2
	(d)	40 UNION, RESTRICTOR	D9798		1
		40 UNION, RESTRICTOR	1-01288		1
		40 UNION, RESTRICTOR	69-64545-1		1
		40 UNION, RESTRICTOR	69-64545-4		1
		40 UNION, RESTRICTOR	1-01188		1
		40 UNION, RESTRICTOR	118-10398		1
		41 O-RING	BACP11PB6		1
	(e)	NO PARTS REQUIRED			
	(f)	42 TUBE ASSEMBLY	65-58135-36		1
	(g)	43 REDUCER	MS21916J6-4		2
		44 O-RING	BACP11PB4		2

Oil Pressure Transmitter and Switch Installation
Figure 543 (Sheet 5)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
X. (h)	45	TUBE ASSEMBLY	65-58135-32		1
	46	TUBE ASSEMBLY	65-58135-38		1
(i)		NO PARTS REQUIRED			
(j)	-47	CLAMP	BACC10HS06		4
	-48	SCREW	BACS12CB3-7		2
	-49	NUT	BACN10JC3		2
(k)		NO PARTS REQUIRED			

*[1] Preassemble in par. 3.F.

Oil Pressure Transmitter and Switch Installation
Figure 543 (Sheet 6)



OVERHAUL MANUAL

Y. Install engine oil temperature transmitter as follows: (See figure 544.)

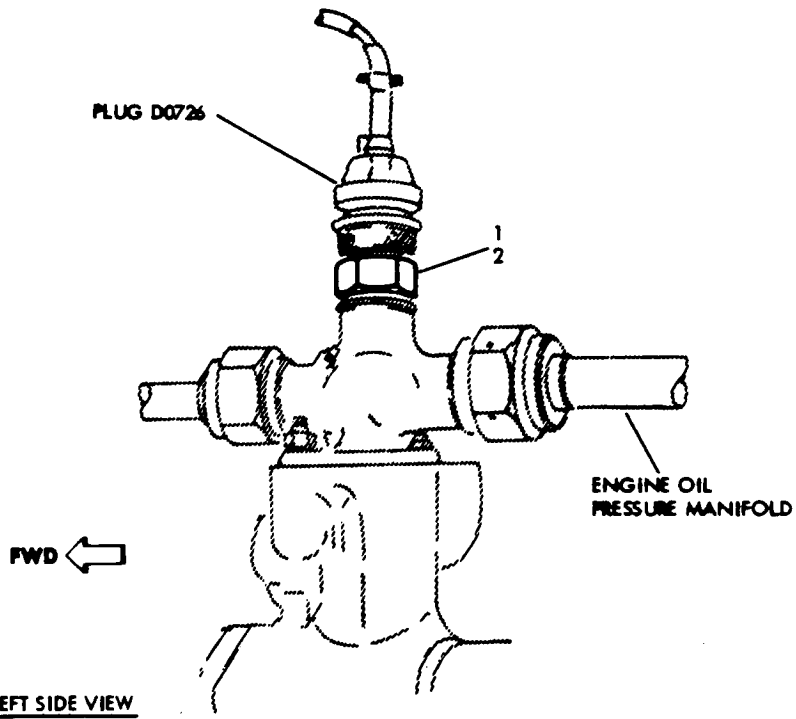
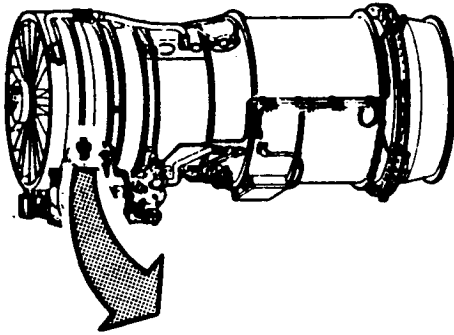
- (1) Install engine oil temperature transmitter (1) and O-ring (2) in oil temperature transmitter port in engine oil pressure manifold.

NOTE: Make electrical bond between transmitter and manifold faying surfaces. Maximum resistance across faying surfaces is 0.0025 ohms (2.5 milliohms).

- (2) Lockwire transmitter in place.

NOTE: Deleted

- (3) Attach plug D0726 to electrical connector in oil temperature transmitter and lockwire in place.



REF.
FIG
1131

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(1)	1	TRANSMITTER, Engine Oil Temperature	56B17C		1
	1	TRANSMITTER, Engine Oil Temperature	56B17D (Optional to 56B17C)		1
	2	O-RING	BACG10U4		1
(2)		No parts required			
(3)		No parts required			

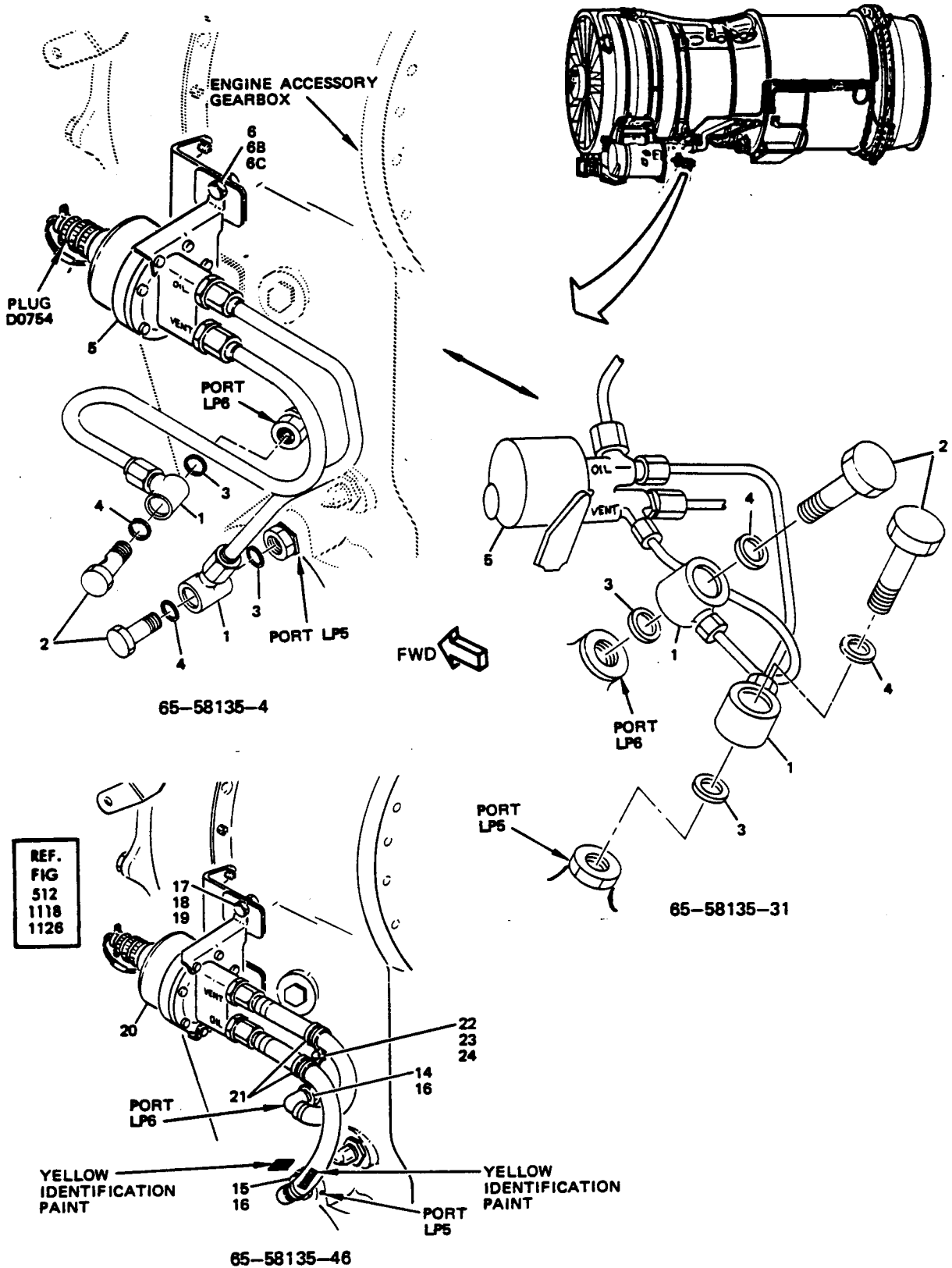
Engine Oil Temperature Transmitter Installation
Figure 544

OVERHAUL MANUAL

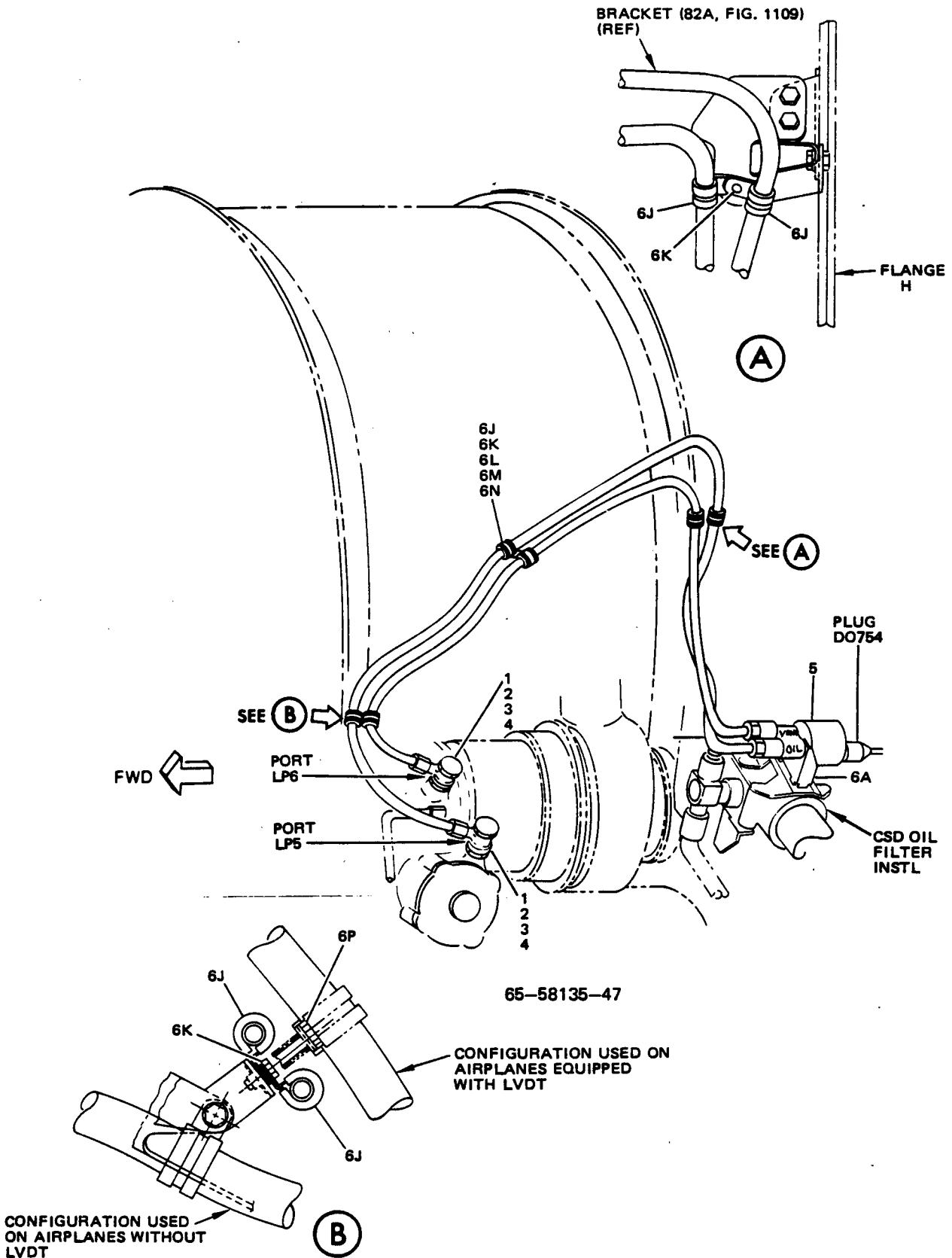
- Z. Install engine oil differential pressure switch as follows: (Fig. 545)
- (1) 65-54476-1, -2, -4 Oil Differential Pressure Switch Installation
- (a) Install universal elbow (1) in oil filter inlet port LP5 with universal bolt (2) and O-rings (3 and 4).
 - (b) Install universal elbow (1) in oil filter outlet port LP6 with universal bolt (2) and O-rings (3 and 4).
 - (c) Locate pressure switch assembly (5) and attach with screws (6, 6A), washers (6B), and nuts (6C). Make electrical bond per 20-11-03 at attaching points. Resistance between mounting base of switch and basic structure shall not exceed 0.100 ohms.
 - (d) 65-58135-4, -31, -47 Plumbing Installation.
 - 1) Attach tube assembly from VENT port of switch to elbow (1) in LP6 port.
 - 2) Attach tube assembly from OIL port of switch to elbow (1) in LP5 port.
 - (e) 65-58135-48 Plumbing Installation
 - 1) Attach tube assembly (7) to elbow (1) in LP5 port.
 - 2) Attach tube assembly (8) to elbow (1) in LP6 port.
 - 3) Connect tube assembly (10) to tube assemblies (12) and (7) using tee (11).
 - 4) Connect tube assembly (9) to tube assemblies (13) and (8) using tee (11).
 - (f) Tighten all end fittings of tube assemblies. Tighten universal bolts (2) to 133-147 lb-in. and secure with lockwire per 20-50-02.
 - (g) Secure tube assemblies with clamps (6J, 6L), using screws (6K, 6P), washer (6M) and nut (6N).
 - (h) Attach electrical plug D0754 to connector on pressure switch.
- (2) 65-54476-3 Oil Differential Pressure Switch Installation
- (a) Install union (15) with O-rings (16) in oil filter inlet LP5 port. Tighten union (15) to 200-225 lb-in.
 - (b) Install union (14) with O-ring (16) in oil filter outlet port LP6. Tighten union (14) to 200-225 lb-in.

- (c) Locate pressure switch assembly (20) and attach with screws (17), washers (18), and nuts (19). Make electrical bond per 20-11-03 at attaching points. Resistance between mounting base of switch and basic structure shall not exceed 0.100 ohms.
- (d) 65-58135-46 Plumbing Installation
 - 1) Attach hose assembly from VENT port of switch to union (14) in LP6 port and tighten.
 - 2) Attach hose assembly from OIL port of switch to union (15) in LP5 port and tighten.
 - 3) Clean and lightly sand area adjacent to LP5 port on gearbox. Apply 0.25-0.30 by 0.50 inch strip of yellow identification paint (SRF-14.9815-302).
 - 4) Clean area near to LP5 port on hose assembly and apply one coat BMS 10-11, type 1 primer (F-20.02). Apply 0.25-0.30 by 0.50 inch strip of yellow identification paint (SRF-14.9815-302).
 - 5) Secure hose assys using clamp (21), screw (22), washer (23) and nut (24).
 - 6) Attach electrical plug D0754 to connector on pressure switch.

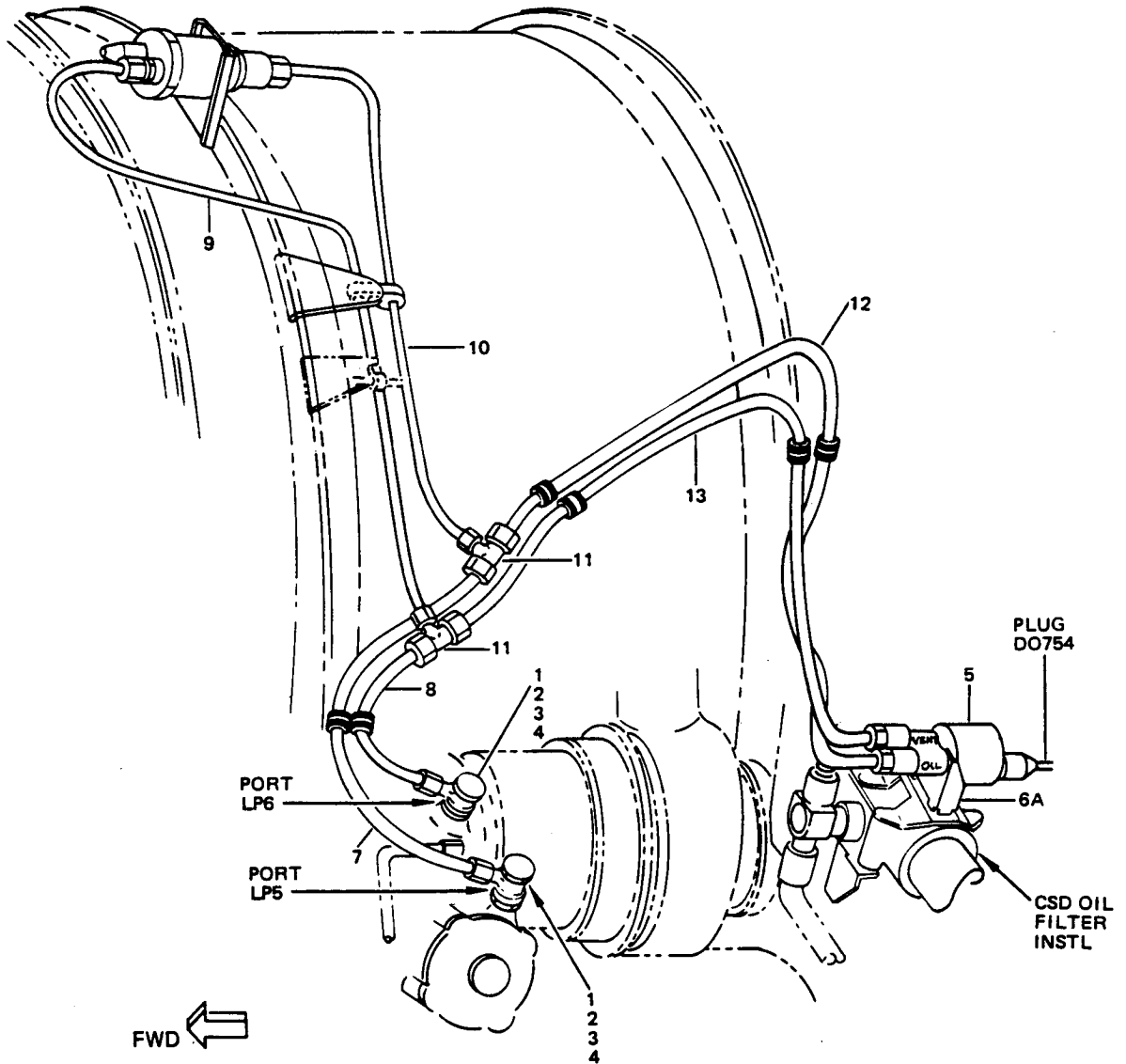
OVERHAUL MANUAL



Engine Oil Differential Pressure Switch Installation
Figure 545 (Sheet 1)



Engine Oil Differential Pressure Switch Installation
Figure 545 (Sheet 2)



65-58135-48

Engine Oil Differential Pressure Switch Installation
Figure 545 (Sheet 3)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
Z. (1)	1	ELBOW, UNIVERSAL	118-10557		2	
	2	BOLT, UNIVERSAL FITTING	NAS551-4H		2	
	3	O-RING	BACP11P-B4		2	
	4	O-RING	BACP11P11		2	
	5	SWITCH ASSEMBLY	(PREASSEMBLED IN PAR. 3.K.)		RF	
	6	SCREW	NAS1801-3-8		2	
	6A	SCREW	NAS1801-3-7		2	
	6B	WASHER	AN960-10		2	
	6C	NUT	NAS679A3W		2	
	6J	CLAMP	BACC10HS06		AR	
	6K	SCREW	NAS1801-3-9		3	
	6L	CLAMP	BACC10HS20		1	
	6M	WASHER	AN960D10L		1	
	6N	NUT	BACN10JC3		1	
	6P	SCREW	NAS1801-3-22		AR	
	7	TUBE ASSEMBLY	65-58135-40		1	
	8	TUBE ASSEMBLY	65-58135-43		1	
	9	TUBE ASSEMBLY	65-58135-44		1	
	10	TUBE ASSEMBLY	65-58135-41		1	
	11	TEE	MS21905-6		2	
	12	TUBE ASSEMBLY	65-58135-42		1	
	13	TUBE ASSEMBLY	65-58135-45		1	
	(2)	14	UNION	69-72689-1		1
		15	UNION	69-72689-2		1
16		O-RING	BACP11PB4		2	
17		SCREW	NAS1801-3-8		2	
18		WASHER	AN960-10		2	
19		NUT	NAS679A3W		2	
20		SWITCH ASSEMBLY	(PREASSEMBLED IN PAR. 3.K.)			
21		CLAMP	BACC10HS11		1	
22		SCREW	BACS12CB3-8		1	
23		WASHER	AN960D10L		1	
24		NUT	BACN10JC3		1	

AA. Install engine starter as follows: (Fig. 546)

- (1) Lubricate input spline on starter shaft with grease, BMS 3-24, prior to installation.
- (2) Locate starter (1) and gasket (2) on engine starter pad. Mesh starter spline with engine output spline and secure starter with nuts (3).

NOTE 1: Electrically bond nuts (3) to starter per 20-11-03.

NOTE 2: The nut at the 3 o'clock position is not installed because of limited accessibility.

- (3) Attach retainer (4) to shutoff valve (5) with coupling (6).
- (4) Attach shutoff valve (5) to starter (1) with coupling (7).

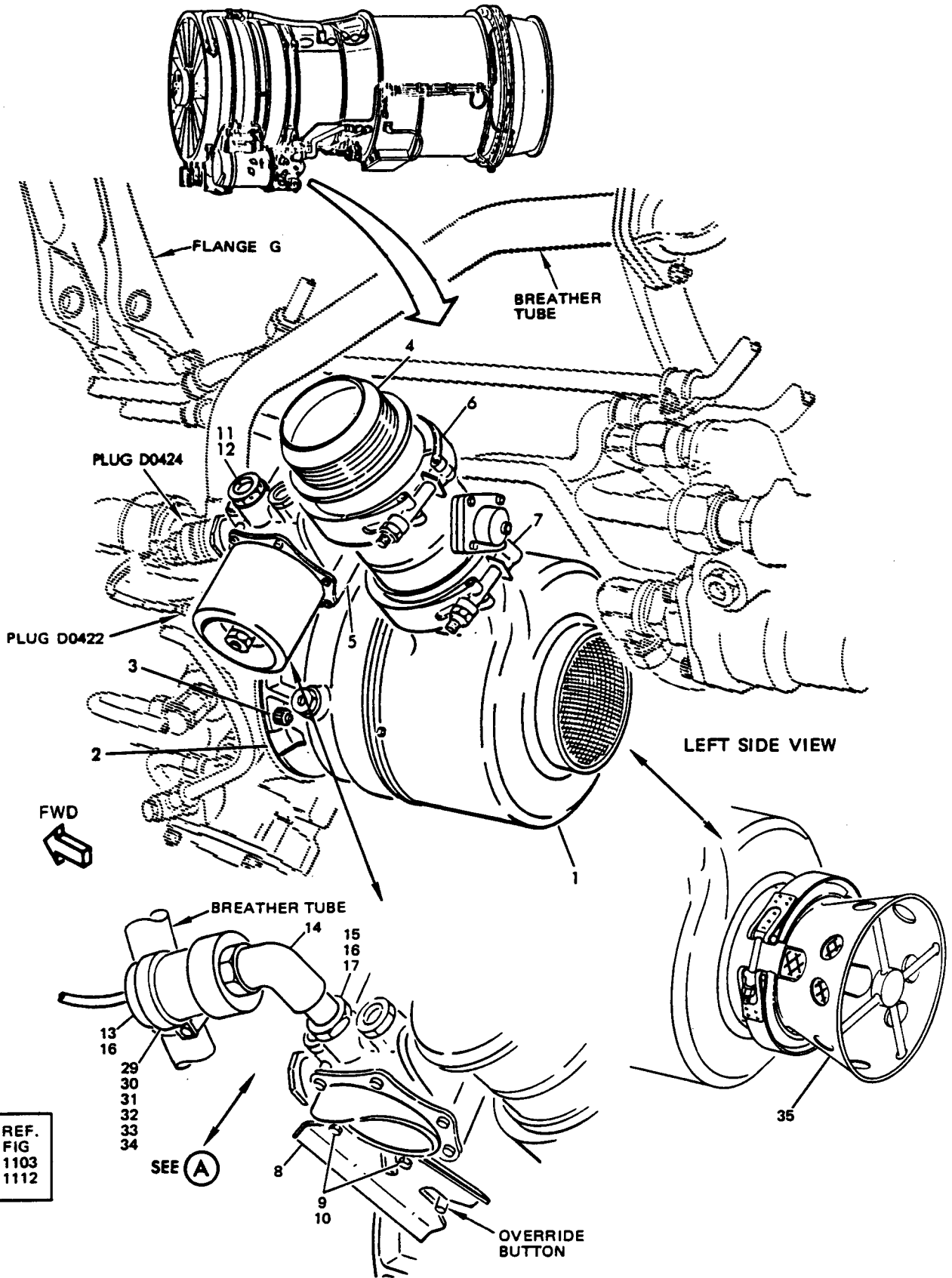
NOTE: Maintain 0.20- to 0.30-inch clearance between valve electrical connector boss and breather tube with valve P/N 10-60706-2, -3 and -4 and 0.40- to 0.60-inch clearance with valve P/N 10-60706-5.

- (5) To install guide (8), remove and discard two existing fasteners from starter (1). Locate guide (8) and secure with screws (9) and washers (10). Tighten screws (9) to 25-30 pound-inches and secure with lockwire.

NOTE: Remove lockwire (if used) from override button behind guide.

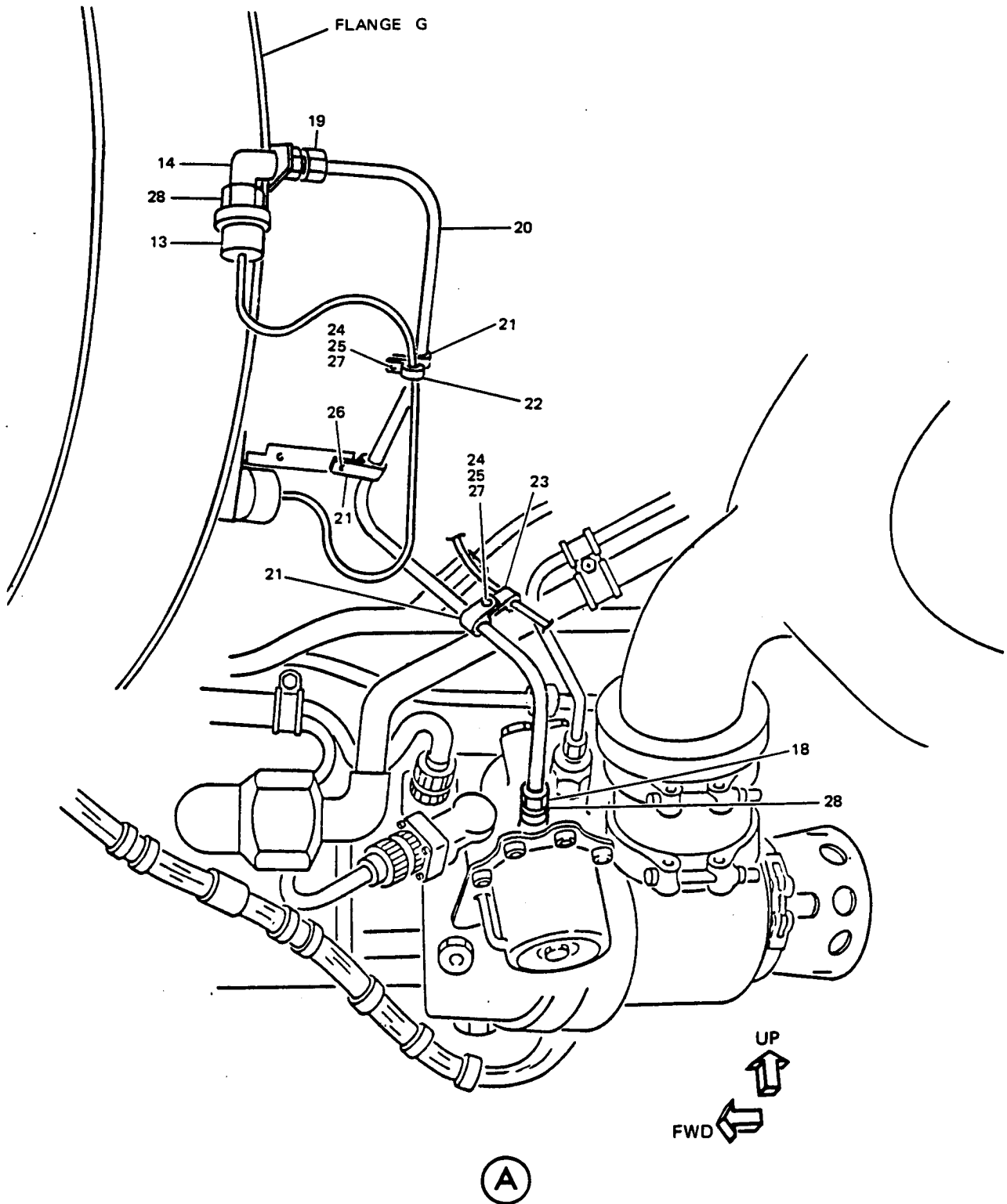
- (6) Installations without pressure switch -- Install plug and bleeder (11) with O-ring (12), and secure with lockwire.
- (7) Install engine starter valve open indication light pressure switch.
 - (a) Starter mounted switch -- Install pressure switch (13) on starter using 90° elbow (14), O-rings (16), washers (17), and nut (15). Position pressure switch as necessary to attain 0.30-0.40 inch clearance between pressure switch and breather tube.
 - (b) Flange mounted switch -- Install pressure switch (13) on flange G using 90° elbow (14), packing (28), and nut (19). Install union (18) and packing (28) in start valve. Connect pressure switch (13) to start valve with tube assembly (20). Secure tube assembly (20) with clamps (21, 22, 23), washers (24), screws (25, 26), and nuts (27). Connect pressure switch (13) to breather tube using clamps (33, 34) and fasteners (29, 30, 31, 32).

- (8) Install cover on open end of retainer (4).
- (9) Mate plugs D0424 and D0422 to electrical connectors on starter.
- (10) Install containment assembly.
 - (a) Containment assembly 65C13014-1 -- Install on 383152-1-2 or 383152-16-1 starter using coupling supplied as part of starter. Tighten coupling bolt to 25-40 lb-in and secure with lockwire per 20-50-02.
 - (b) Containment assembly 65C13014-3 --Install on 383152-19-1 starter using coupling (30). Tighten coupling bolt to 45-55 lb-in. and secure with lockwire per 20-50-02.



REF.
FIG.
1103
1112

Engine Starter Installation
Figure 546 (Sheet 1)



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AA.					
(1)	1	NO PARTS REQUIRED			
(2)	1	STARTER	383152-16-1 *[6]		1
	1	STARTER	383152-1-2		1
	1	STARTER	383152-19-1 *[7]		1
	1	STARTER	383342-1-1		1
	2	GASKET	AN4047-1		1
	2	GASKET	MS9136-01		1
	3	NUT	BACN10BL6L		5
	3	NUT	BACN10GW6 (OPTIONAL)		5
(3)	4	RETAINER (SEE DUCT INSTL)	69-20823-1		1
	5	VALVE, SHUTOFF	10-60706-8 *[5]		1
	5	VALVE, SHUTOFF	10-60706-5		1
	5	VALVE, SHUTOFF	10-60706-4 *[1]		1
	5	VALVE, SHUTOFF	10-60706-3 *[1]		1
	5	VALVE, SHUTOFF	10-60706-2 *[1]		1
	6	COUPLING (SEE DUCT INSTL)	BACCL0DP300A		1
(4)	5	VALVE, SHUTOFF	10-60706-8 *[5]		1
	5	VALVE, SHUTOFF	10-60706-5		1
	5	VALVE, SHUTOFF	10-60706-4 *[1]		1
	5	VALVE, SHUTOFF	10-60706-3 *[1]		1
	5	VALVE, SHUTOFF	10-60706-2 *[1]		1
	7	COUPLING	BACCL0DP300A		1
	(5)	8	GUIDE	66-20269-800 (PREF)	
8		GUIDE	66-20269-2 *[1]		1
8		GUIDE	66-20269-1 *[1]		1
9		SCREW	MS24673-3		2
10		WASHER	BACW10BP3APU *[2]		2
10		WASHER	AN960C10 *[3]		2
10		WASHER	AN960PD10 *[4]		2
(6)	11	PLUG AND BLEEDER	AN814-6SL		1
	12	O-RING	BACP11P-B6		1
(7)	13	PRESSURE SWITCH	10-60927-1		1
	13	PRESSURE SWITCH	10-60927-2		1
	13	PRESSURE SWITCH	10-60927-3		1
	14	ELBOW	2-01557		1
	15	NUT	BACN10DU6		1
	16	O-RING	BACP11PB6A		2

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
	17	WASHER	MS28777-6		1
	18	UNION	MS21902J6		1
	19	NUT	BACN10DR6J		1
	20	TUBE ASSY	65-58126-26 *[9]		1
	20	TUBE ASSY	65-58126-27 *[10]		1
	21	CLAMP	BACC10HS06		3
	22	CLAMP	BACC10GE3C		1
	23	CLAMP	BACC10GE4C		1
	24	WASHER	AN960C10L		2
	25	SCREW	NAS1802-3-6		2
	26	SCREW	NAS1802-3-10		1
	27	NUT	BACN10JC3CM		2
	28	PACKING	BACP11K6 *[8]		2
	29	SCREW	NAS1801-3-24		1
	30	NUT	BACN10JC3		1
	31	WASHER	AN960D10		1
	32	SPACER	NAS43HT4-64		1
	33	CLAMP	BACC10HS20		1
	34	CLAMP	BACC10GE22C		1
(10)	35	CONTAINMENT ASSY	65C13014-1 *[6]		1
	35	CONTAINMENT ASSY	65C13014-3 *[7]		1
	36	COUPLING CLAMP	BACC10GY-400		1

- | | |
|----------------------------|---|
| *[1] OPTIONAL | *[6] 383152-16-1 USED WITH 65C13014-1 |
| *[2] USE WITH 66-20269-800 | *[7] 383152-19-1 USED WITH 65C13014-3 |
| *[3] USE WITH 66-20269-2 | *[8] USED WITH FLANGE MOUNTED PRESSURE SWITCH |
| *[4] USE WITH 66-20269-1 | *[9] USED WITH PARKER START VALVE |
| *[5] REPLACES 10-60706-5 | *[10] USED WITH AIRESEARCH START VALVE |

Engine Starter Installation
Figure 546 (Sheet 4)

AB. Install cabin air and engine starter ducts as follows: (Fig. 547)

- (1) Install manifold assembly (1) on left-hand 8th-stage bleed pad with bolts (2, 3, 4) and washers (5). Use gasket supplied with bleed port cover.

NOTE: A maximum gap of 0.050 inch is allowed between surfaces of manifold flange and bleed pad before installation of bolts.

- (2) Lockwire bolts (2, 3, 4) per 20-50-02.

- (3) Attach upper and lower link assemblies to existing brackets on flange "H", using spacers (7), bolts (8), washers (9), and nuts (10). To eliminate preload on duct, adjust lengths of link assemblies (bolt (8) center to inside surface of bracket) to 2.73-2.93 inches for upper link and 1.020-1.220 inches for lower link, and tighten jam nuts. Should either link length be outside the specified limits, check duct assembly alignment.

- (4) Assemble rod end bearings (11, 12) and jamnut (13). Adjust length (center to center of rod end bearings) to 2.19-2.39 inches, tighten jamnut (13) finger tight.

- (5) Attach rod end bearing (12) to duct assembly (14) with bolt (15), washer (16), and nut (17).

- (6) Locate duct assembly (14 or 14A) for preliminary attachment to manifold assembly (1) and engine starter, and fit wear rings and shims as follows: (Details C and D)

NOTE: Align starter scroll to duct assembly (14 or 14A) by loosening the eight bolts that attach scroll to starter body. Rotate scroll inlet to desired position and retighten bolts to 34-45 lb-in.

- (a) Install wear ring (19) in bearing nut which attaches to starter. If duct assembly (14A) is used, install wear ring (18) in bearing nut which attaches to manifold assembly (1).

- (b) Attach bearing nuts to starter and manifold assembly and tighten to 70-90 lb-in.

- (c) Measure gap between bearing nut and retainer in each location with a feeler gage (Gap X in detail C and D).

- (d) Add 0.012-0.014 inch to each measurement to obtain thickness of shims to be used in each location for final installation of duct assembly.

- (e) Adjust thickness of shims (22, 20), if used, by removing 0.003-inch laminations as necessary.

NOTE: When optional shims (20, 21, 22 and 23) are used, coat delaminated surface with BMS 10-11, type 1 primer prior to installation.

- (f) Detach duct assembly from starter and manifold assembly (1).
- (g) Coat threads of bearing nuts and retainers with anti-seize compound, MIL-A-907. Reinstall duct assembly (14 or 14A) with wear rings (18, 19) and shims (20, 21, 22, 23) in place.

NOTE: Edges of shims must be flush with outer surface of bearing nuts.

- (h) Tighten bearing nuts to 380-400 lb-in.
 - (i) Check joint for freedom of movement. Lockwire bearing nut to retainer per 20-50-02.
- (7) Attach aft end of duct assembly (14 or 14A) to left side of precooler assembly with clamp (24).
- (8) Attach rod end bearing (11) to existing bracket on flange "J1" with spacer (25), bolt (26), washer (27), and nut (28). To eliminate preload on duct, check that link length (center to center of attaching bolts) is 2.19-2.39 inches, and tighten jamnut (13).

NOTE: Clearance between duct assembly (14A) and upper aft mount bracket on engine exciter box shall be 0.15 inch minimum.

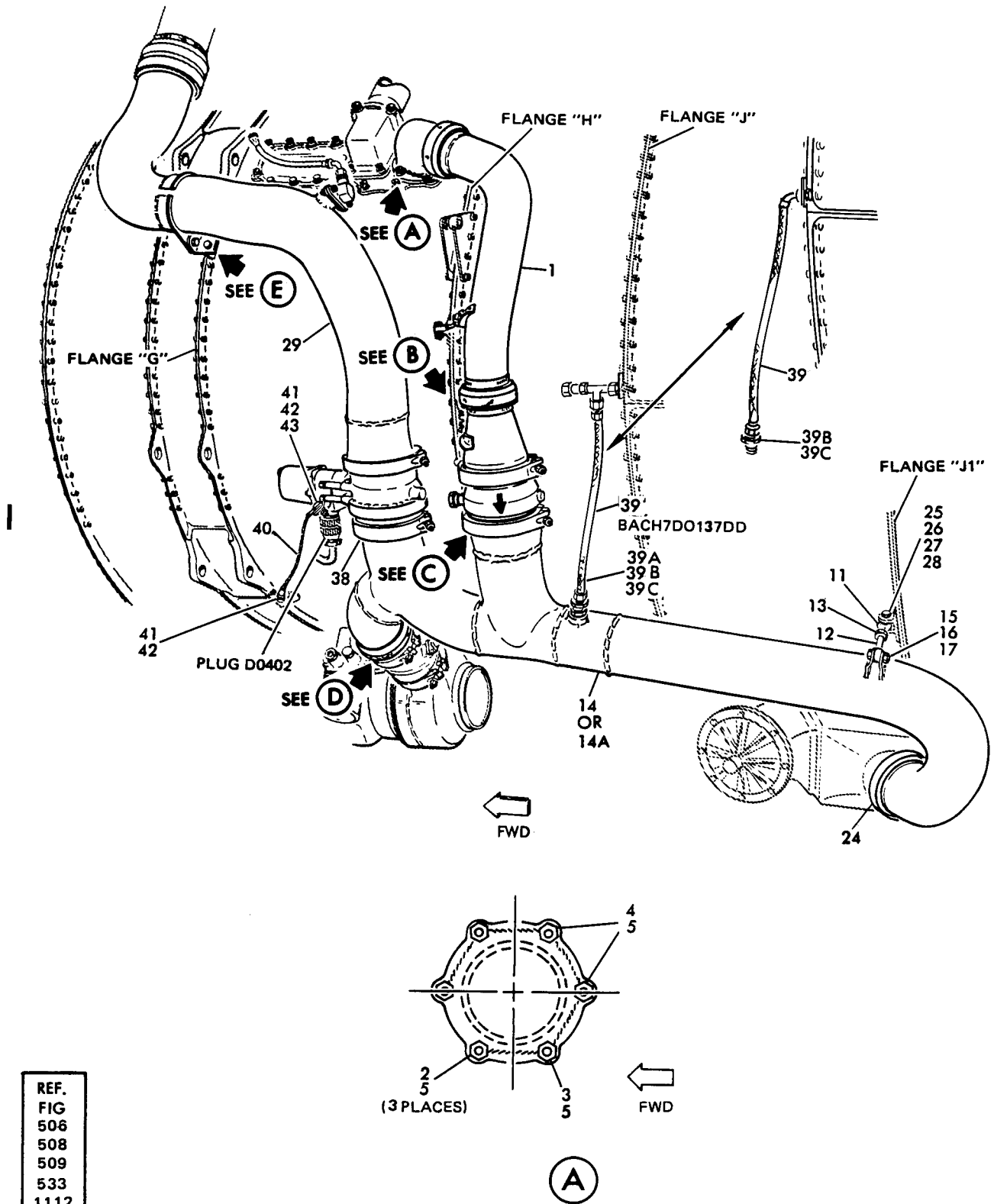
- (9) Attach clips (30, 31) to manifold assembly (29) with screws (32), spacers (33), and washers (34).
- (10) Install manifold assembly (29) by attaching to existing bracket on flange "G" with bolts (35), washers (36), and nuts (37).

CAUTION: MANIFOLD MUST CLEAR ENGINE MOUNT BY MINIMUM OF 0.250 INCHES.

- (11) Attach valve on end of manifold assembly (29) to duct assembly (14 or 14A) with clamp (38).
- (12) Attach plug D0402 to electrical connector on bottom of valve on manifold assembly (29). Lockwire plug in place.
- (13) Install hose assembly (39) with reducer (39A), packing (39B) and union (39C) between tee on flange "J" (installed in par. 5.N.) and fitting in welded boss on starter duct. Tighten end fittings of hose.

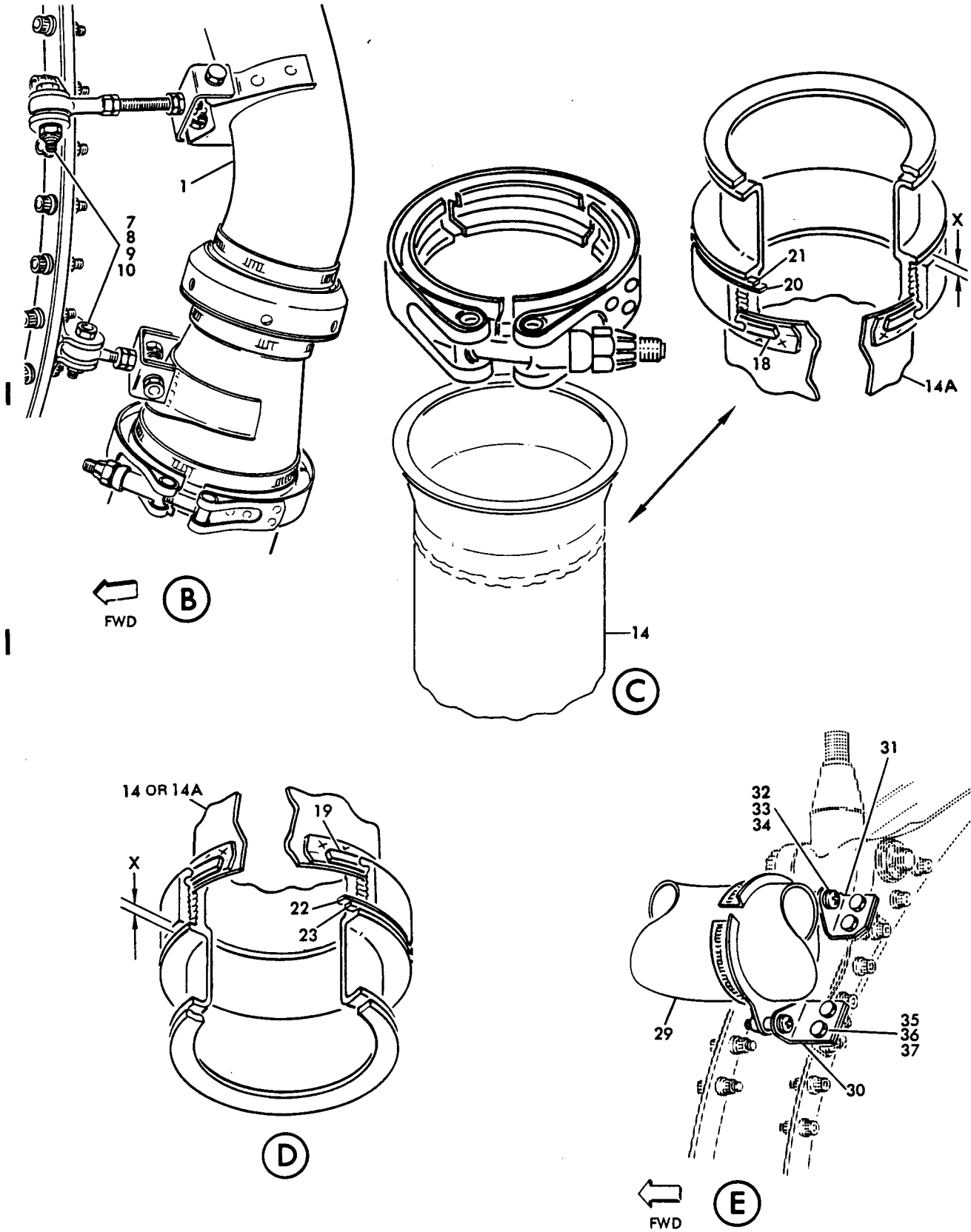
NOTE: When optional hose assembly (39) is used, install with packing (39B) and union (39C).

- (14) Install bonding jumper (40) between bonding tab on valve assembly and bracket on flange "G" with screws (41), washers (42), and nuts (43). Tighten nuts (43) to 20-30 lb-in. Maximum allowable resistance between bonding jumper and structure is 0.001 ohms (1 milliohm).



REF.
FIG
506
508
509
533
1112


Cabin Air and Engine Starter Ducts Installation
Figure 547 (Sheet 1)




Cabin Air and Engine Starter Ducts Installation
 Figure 547 (Sheet 2)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AB. (1)	1	MANIFOLD ASSY	(PREASSEMBLED IN PARA. 3.H. (2))		RF
	2	BOLT	BACB30MT4HT2		3
	3	BOLT	BACB30MT4HT6		1
	4	BOLT	BACB30MT4HT7		2
	5	WASHER	BACW10BP4CTU		6
(2)		NO PARTS REQUIRED			
(3)	7	SPACER	NAS76-4-005P		2
	8	BOLT	NAS1104-11		2
	9	WASHER	AN960C416		2
	10	NUT	BACN10JC4		1
(4)	11	BEARING, ROD END	BACB10J12		1
	12	BEARING, ROD END	BACB10A192M2		1
	13	JAMNUT	AN316-4R		1
(5)	14	DUCT ASSY	65-53863-32		1
	14	DUCT ASSY	65-53863-28 (OPT)		1
	14A	DUCT ASSY	65-53863-22 (OPT)		1
	14A	DUCT ASSY	65-53863-1 (OPT)		1
	15	BOLT	NAS1104-9		1
	16	WASHER	AN960-416		1
	17	NUT	BACN10JC4		1
(6)	18	RING, WEAR	69-26229-2		1
	19	RING, WEAR	69-26230-2		1
	20	SHIM, LAMINATED	69-56508-3		1
	21	SHIM, SOLID	69-56508-6		1
	22	SHIM, LAMINATED	69-56508-2		1
	23	SHIM, SOLID	69-56508-5		1
(7)	24	CLAMP	BACC10DU350Y		1

Cabin Air and Engine Starter Ducts Installation
Figure 547 (Sheet 3)

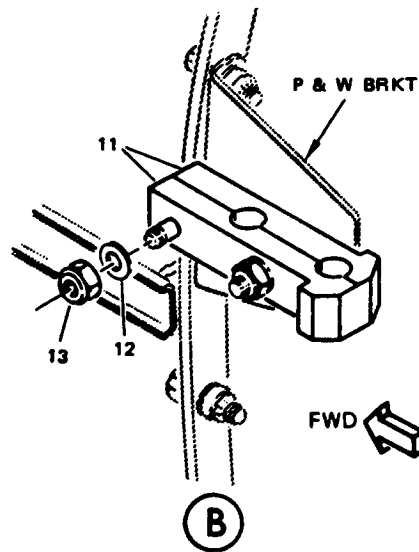
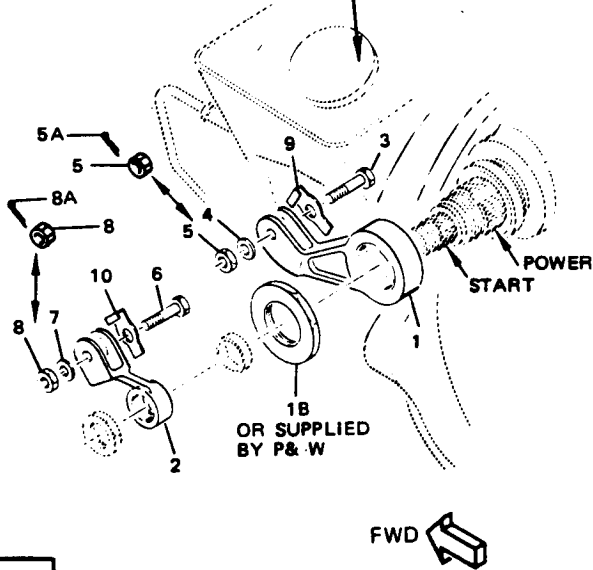
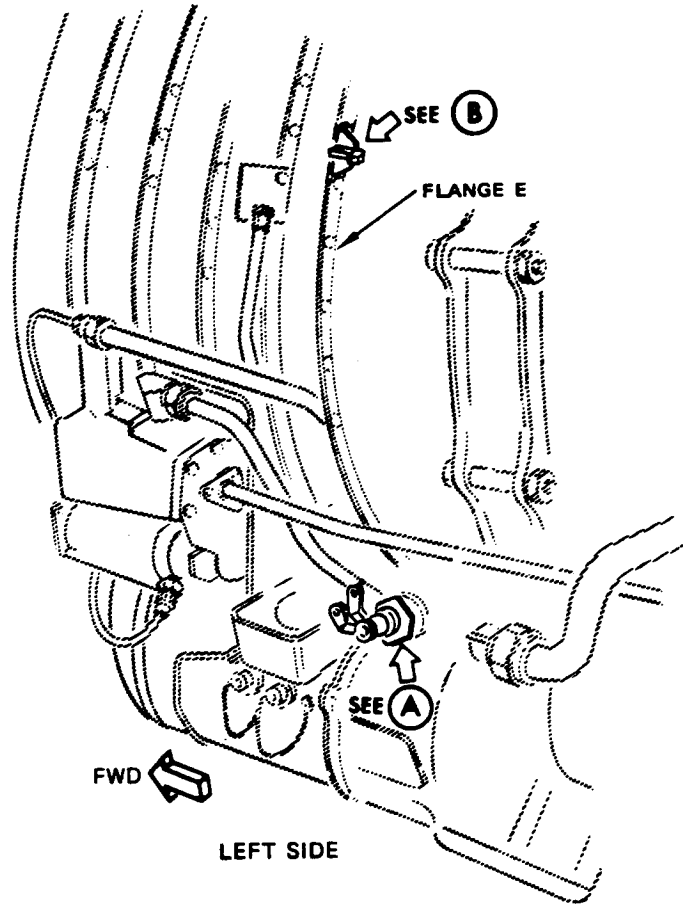
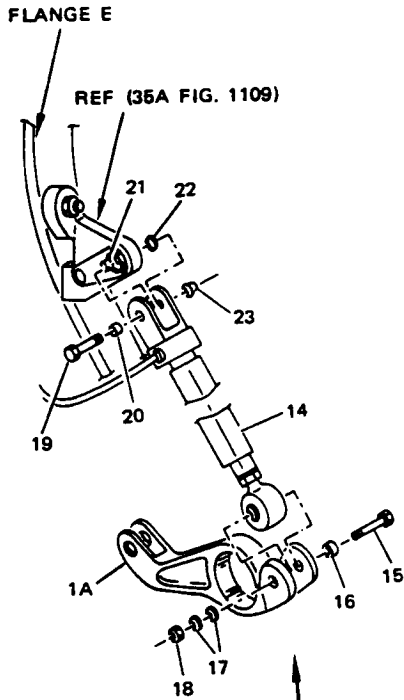
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
(8)	25	SPACER	NAS76-4-005P		1
	26	BOLT	NAS1104-11		1
	27	WASHER	AN960C416		1
	28	NUT	BACN10JC4		1
(9)	29	MANIFOLD ASSY	(PREASSEMBLY IN PAR. 3.H.(1))		RF
	30	CLIP	69-43424-5		1
	31	CLIP	69-43424-6		1
	32	SCREW	NAS6603-2		2
	33	SPACER	NAS43HT3-8		2
	34	WASHER	AN960C10		2
(10)	35	BOLT	NAS1103-1		4
	36	WASHER	AN960-10L		4
	37	NUT	BACN10JC3		4
(11)	38	CLAMP	BACC10DU350Y		1
(12)		NO PARTS REQUIRED			
(13)	39	HOSE ASSY	BACH7D0137DD		1
	39	HOSE ASSY 	16236		1
	39A	REDUCER	118-10426		1
	39B	PACKING	BACP11PB6		1
	39C	UNION	MS21902-6		1
(14)	40	JUMPER BONDING	MS25083-2BB10		1
	41	SCREW	BACS12CB3-8		2
	42	WASHER	AN960-10		2
	43	NUT	BACN10JC3		2

 OPTIONAL PART

Cabin Air and Engine Starter Ducts Installation
 Figure 547 (Sheet 4)

AC. Install engine control linkage as follows (Fig. 548):

- (1) Remove P&W nuts and spacers from cross-shafts.
- (2) Install power crank (1) or crank assy (1A) using nut (1B) with crank assy (1A) or nut supplied by P&W with crank (1).
- (3) Install short P&W spacer (0.428 inch long) on cross-shaft.
- (4) Install starter crank (2), secure with P&W nut and lockwire.
- (5) Loosely install items (3 thru 13) as shown. Do not bend tabs of bolt retainer (9, 10) at this time. Final installation of items (3 thru 13), control assys, rub strips and clamps will be accomplished when engine is installed on airplane.
- (6) 65-51534-8 Installation
 - (a) Assemble bushings (16, 20) on bolts (15, 19) using thin coat of corrosion preventive compound (F-19.05).
 - (b) Install stop (21) on bracket assy (35A, Fig. 1109) by sliding stop (21) over spacer and rivet on bracket assy.
 - (c) Install transducer (14) using bolt (15), bushing (16), washers (17) and nut (18) at crank assy end, and bolt (19), bushing (20) washer (22) and nut (23) at bracket assy end.
 - (d) Connect plug on transducer to electrical connector D1831 on flange D (Fig. 525).



REF.
 FIG
 536
 1110

(A)

(B)

Engine Control Linkage Installation
 Figure 548 (Sheet 1)

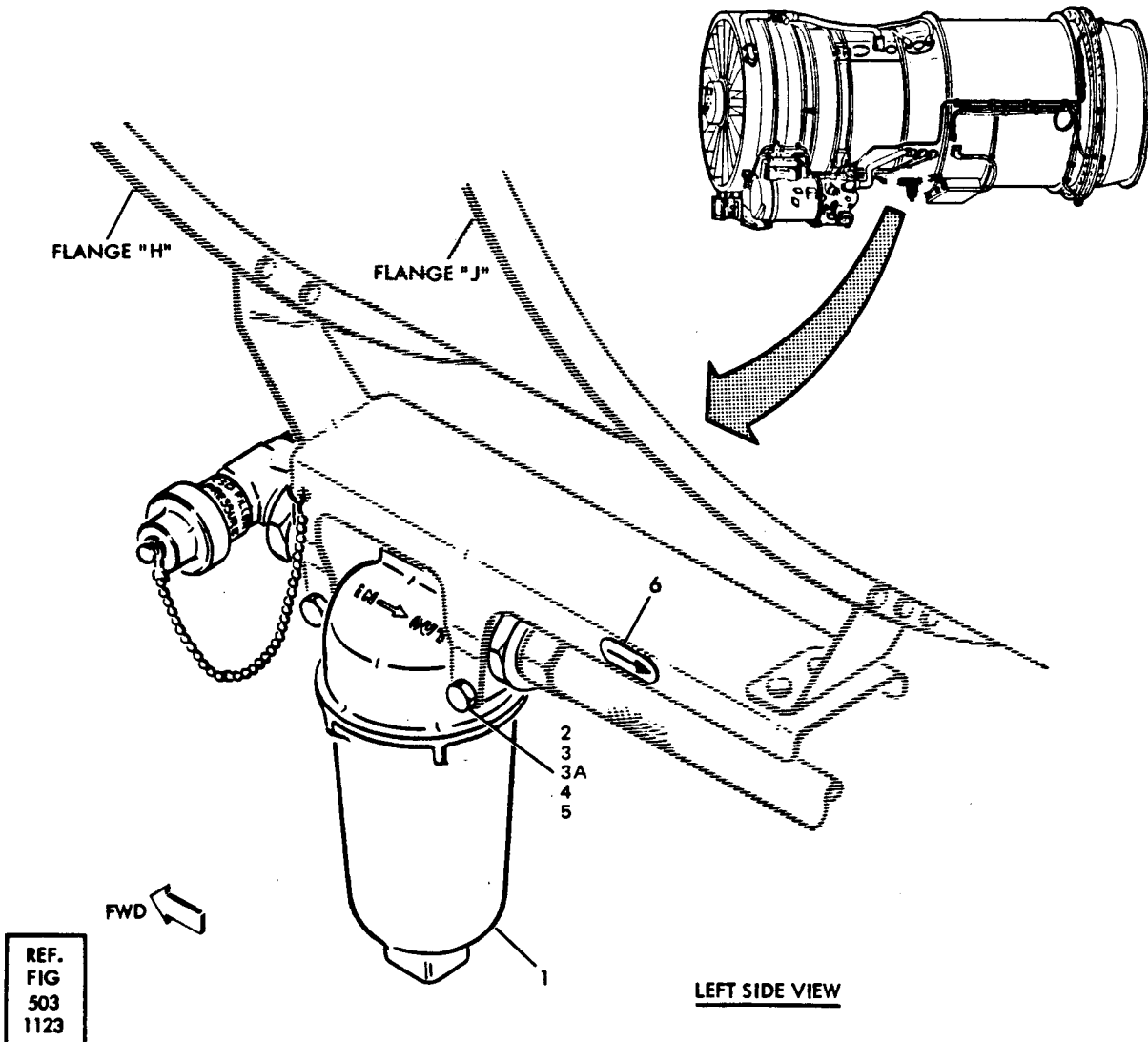
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AC. (1)		NO PARTS REQUIRED			1
(2)	1	CRANK, ENGINE THRUST	65-50546-1		1
	1A	CRANK ASSY, ENGINE THRUST	65-51534-10		1
	1A	CRANK ASSY, ENGINE THRUST	65-51534-11 (OPT)		1
	1B	NUT (USED WITH 1A)	69-71542-1		1
(3)		NO PARTS REQUIRED			
(4)	2	CRANK, ENGINE START	65-50545-3		1
	2	CRANK, ENGINE START	65-50545-1 (OPT)		1
(5)	3	BOLT	BACB3ONE4-11		1
	3	BOLT	BACB3ONF4D11		1
	3	BOLT	NAS1104-11		1
	4	WASHER	AN960C416		1
	5	NUT	BACN10JC4		1
	5	NUT	BACN10JD104		1
	5A	COTTER PIN	MS24665-138		1
	6	BOLT	BACB3ONE5-11		1
	6	BOLT	BACB3ONF5D10		1
	6	BOLT	NAS1105-10		1
	7	WASHER	AN960C516		1
	8	NUT	BACN10JC5		1
	8	NUT	BACN10JD105		1
	8A	COTTER PIN	MS24665-138		1
	9	RETAINER, BOLT	69-66647-3		1
	10	RETAINER, BOLT	69-66647-8		1
	11	CLAMP HALF	69-43481-2		2
	12	WASHER	AN960C416		2
	13	NUT	BACN10JC4		2
(6)	14	TRANSDUCER	L6901		1
	14	TRANSDUCER ASSY	65-51534-12		
	15	BOLT	NAS6604-16		1
	16	BUSHING	BACB28AK04-030		1
	17	WASHER	AN960C416		2
	18	NUT	BACN10JC4		1
	19	BOLT	BACB3ONF4-13		1
	20	BUSHING	BACB28AKC4-030		1
	21	STOP	69-71535-1		1
	22	WASHER	AN960C416		1
	23	NUT	BACN10JC4		1

AD. Install CSD oil filter assembly as follows (Fig. 549):

- (1) Locate oil filter assembly (1) between flanges of existing bracket on lower left side of engine, between flanges H and J. Attach filter assembly with bolts (2), spacers (3), washers (4), and nuts (5).
- (2) Attach ring of chain on filter assembly dust cap to hole in flange of bracket as shown, and crimp ring to fasten.

CAUTION: ENSURE THAT METAL-CAL IS INSTALLED WITH ARROW POINTING IN DIRECTION OF FLOW.

- (3) Install Metal-Cal (6) on bracket per 20-50-05.



CSD Oil Filter Installation
Figure 549 (Sheet 1)

737

POWERPLANT BUILDUP

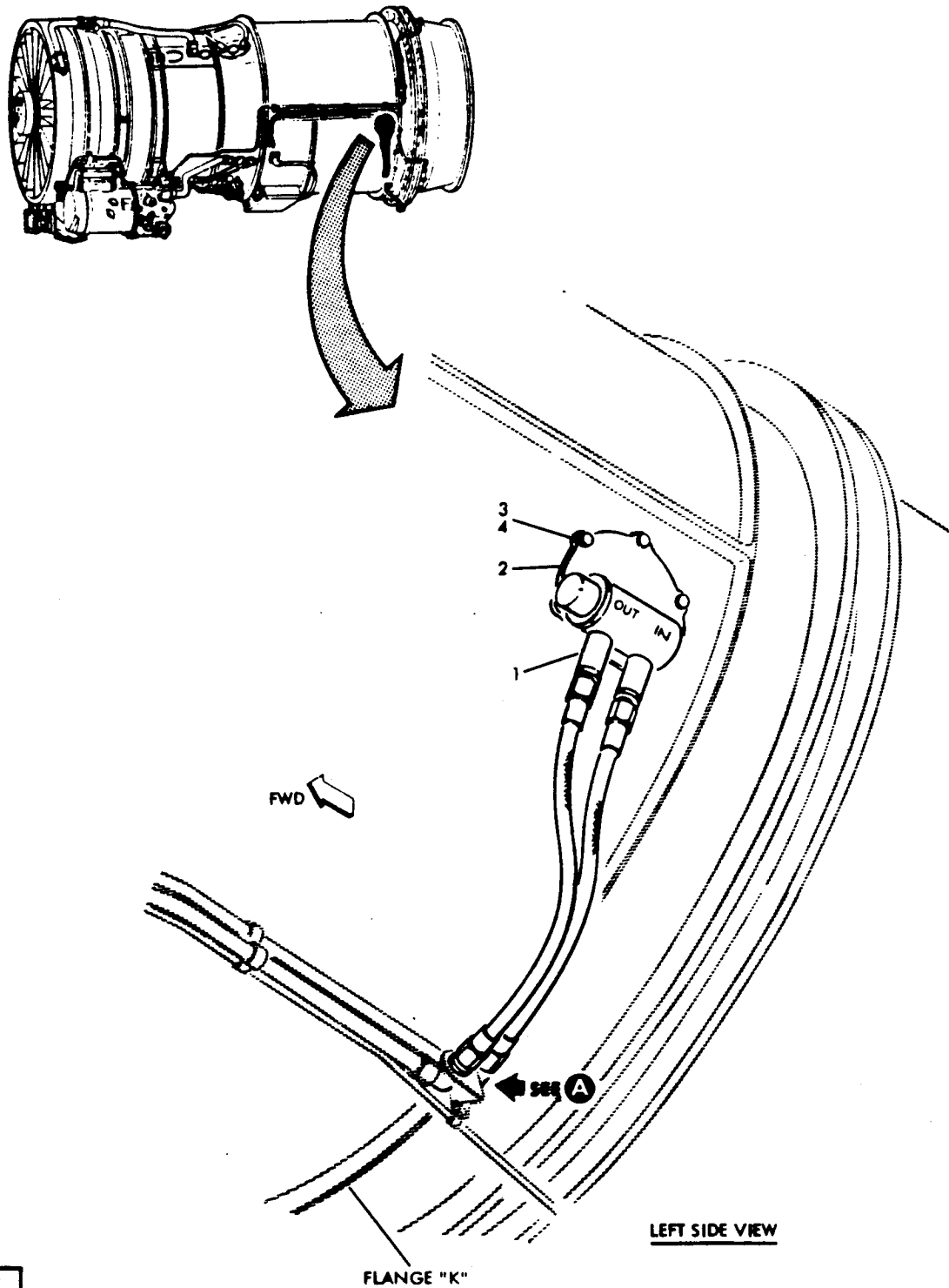


OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AD. (1)	1	FILTER ASSEMBLY, CSD	(PREASSEMBLED IN PAR. 2.D)		RF
	2	OIL			
	2	BOLT	NAS 1104-36		2
	2	BOLT	BACB30NF4-40		2
	3	SPACER	NAS43DD4-28		2
	3A	SPACER	NAS43DD4-16		2
	4	WASHER	AN960C416L		2
	4	WASHER	AN960C416		2
	5	NUT	NAS679A4W		2
	5	NUT	BACN10JC4		2
(2)		NO PARTS REQUIRED			
(3)	6	METAL-CAL	BACM10L001BJZ		1

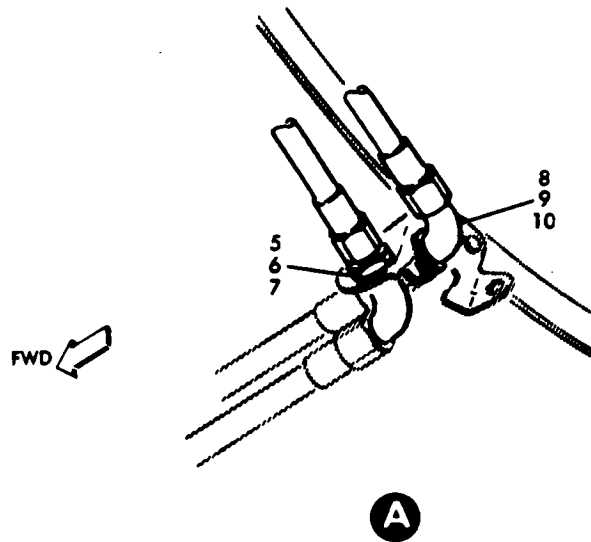
AE. Install CSD oil cooler assembly as follows: (See figure 550.)

- (1) Remove fan bleed pad from left lower side of engine case. Retain nuts and washers to install oil cooler.
- (2) Locate CSD oil cooler assembly (1) on fan bleed pad with gasket (2). Attach parts with washers (3) and nuts (4) furnished with engine.
- (3) Install elbow (5), washer (7), and jamnut (6) on horizontal flange of existing bracket on left side of flange "K." Do not tighten jamnut.
- (4) Install elbow (8), washer (10), and jamnut (9) on vertical flange of existing bracket on left side of flange "K." Do not tighten jamnut.
- (5) Attach free end of hose from IN port of oil cooler to elbow (8). Tighten hose end nut, and tighten jamnut (9).
- (6) Attach free end of hose from OUT port in oil cooler to elbow (5). Tighten end nut of hose and jamnut (6).
- (7) Cap or plug all open ports of fittings.



REF.
FIG
1120

CSD Oil Cooler Installation
Figure 550 (Sheet 1)



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AE. (1)		No parts required			
(2)	1	COOLER ASSEMBLY, CSD Oil	(Preassembled in paragraph 2.E.)		Ref
	2	GASKET	10-61233-4		1
	3	WASHER	AN122582		Ref
	4	NUT	372873		Ref
(3)	5	ELBOW, 90° Bulkhead	MS21908-10		1
	6	JAMNUT	AN924-10		1
	7	WASHER	AN960-1416		1
(4)	8	ELBOW, 90° Bulkhead	MS21908-8		1
	9	JAMNUT	AN924-8		1
	10	WASHER	AN924-8		1
(5)		No parts required			
(6)		No parts required			
(7)		No parts required			

AF. Install aft engine drain fittings as follows: (Fig. 551)

- (1) Attach engine drain fitting assembly (1) and bracket (2) to existing bracket at bottom of flange "K" with screws (3), washers (4), and nuts (5).

NOTE: Make electrical bond between all parts. Maximum allowable resistance between any part and basic structure is 0.0025 ohms (2.5 milliohms) except as noted.

- (2) Attach engine drain fitting assembly (6) to existing bracket at bottom of flange "K" with screws (7), washers (8), and nuts (9).
- (3) Secure wire bundle running aft of drain fittings with clamp (10), screw (11), and nut (12). Tighten nut (12) to 25-35 lb-in.
- (4) Install bracket assembly (13) between existing bracket on flange J1 and bracket (2). Attach with screws (14), washers (15), and nuts (16).
- (5) Complete installation of fire detector as follows:

(a) Sensor P/N 10-61096-115, -315, 10-62061-10

- 1) Install bracket (17) on bracket (13) with screws (18) and nuts (19).
- 2) Connect jumper assemblies (20A, 20B) to fire detector sensor on bracket (17) using nut (20) and washers (20C). Tighten nut (20) to 65-70 lb-in. Make electrical bond check between fire detector sensor terminals and nut (20), maximum allowable resistance is 0.001 ohms (1.0 milliohms).
- 3) Make electrical connection through bracket (17) per electrical installation Fig. 525. Install nuts (20) and lockwire in place.

CAUTION: TO PREVENT TWISTING SENSOR, CHECK THAT SENSOR LOCKNUT IS SECURELY TIGHTENED TO MOUNTING BRACKET AND LOCKWIRED BEFORE ATTACHING WIRE BUNDLE.

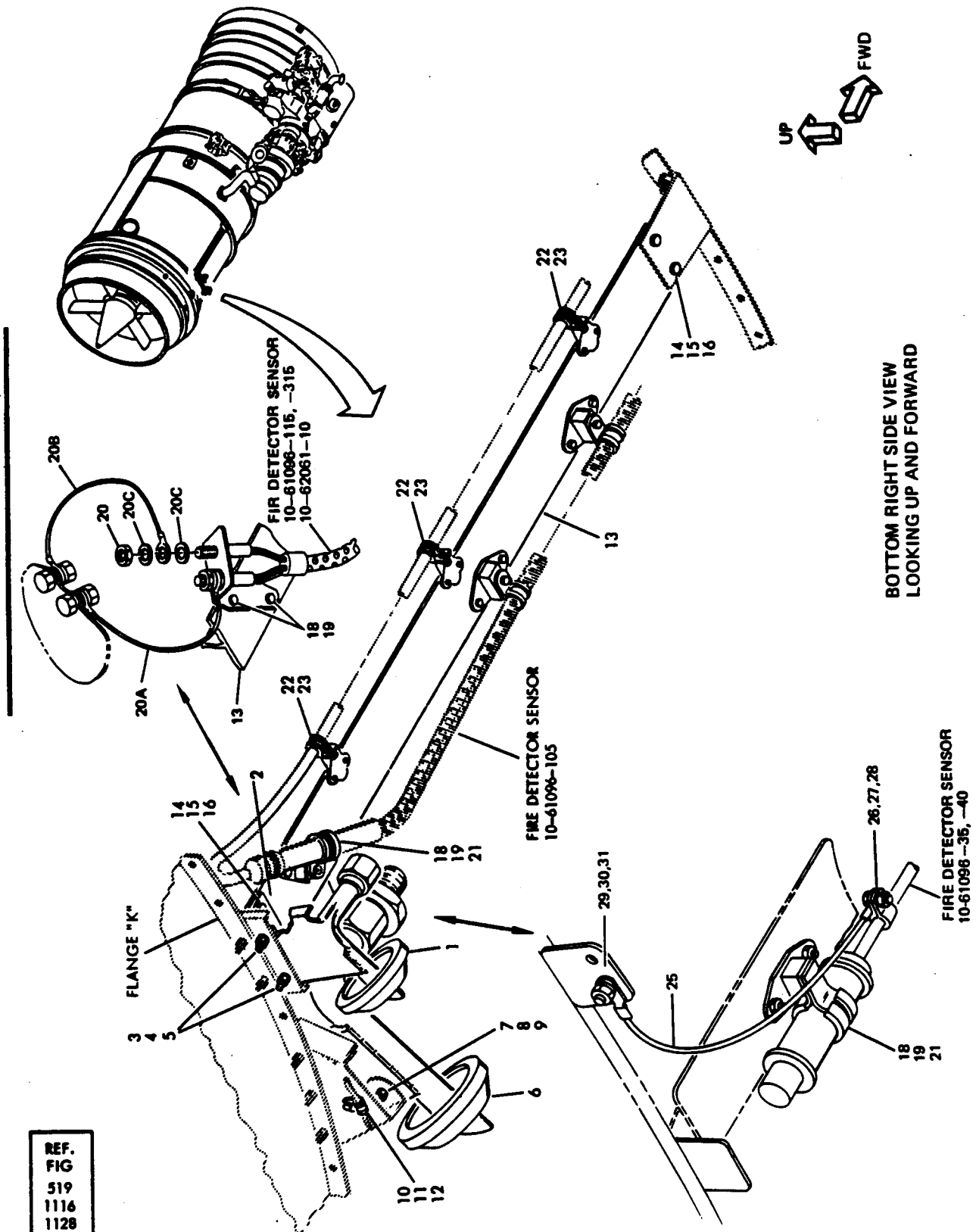
(b) Sensors, P/N 10-61096-105, -35, -40:

- 1) Install quick-release clamp (21) on bracket (13) with screws (18) and nuts (19).
- 2) Make electrical connections per electrical installation Fig. 525.

(c) Sensors, P/N 10-61096-35,-40:

- 1) Install bonding jumper (25) with items (26 thru 31).
- 2) Tighten nut (28) to 20-30 lb-in. Tighten nut (31) to 65-70 lb-in. Maximum allowable resistance is 0.001 ohm (1 milliohm).

(d) Install clamps (22) with screws (23).



REF. FIG
519
1116
1128

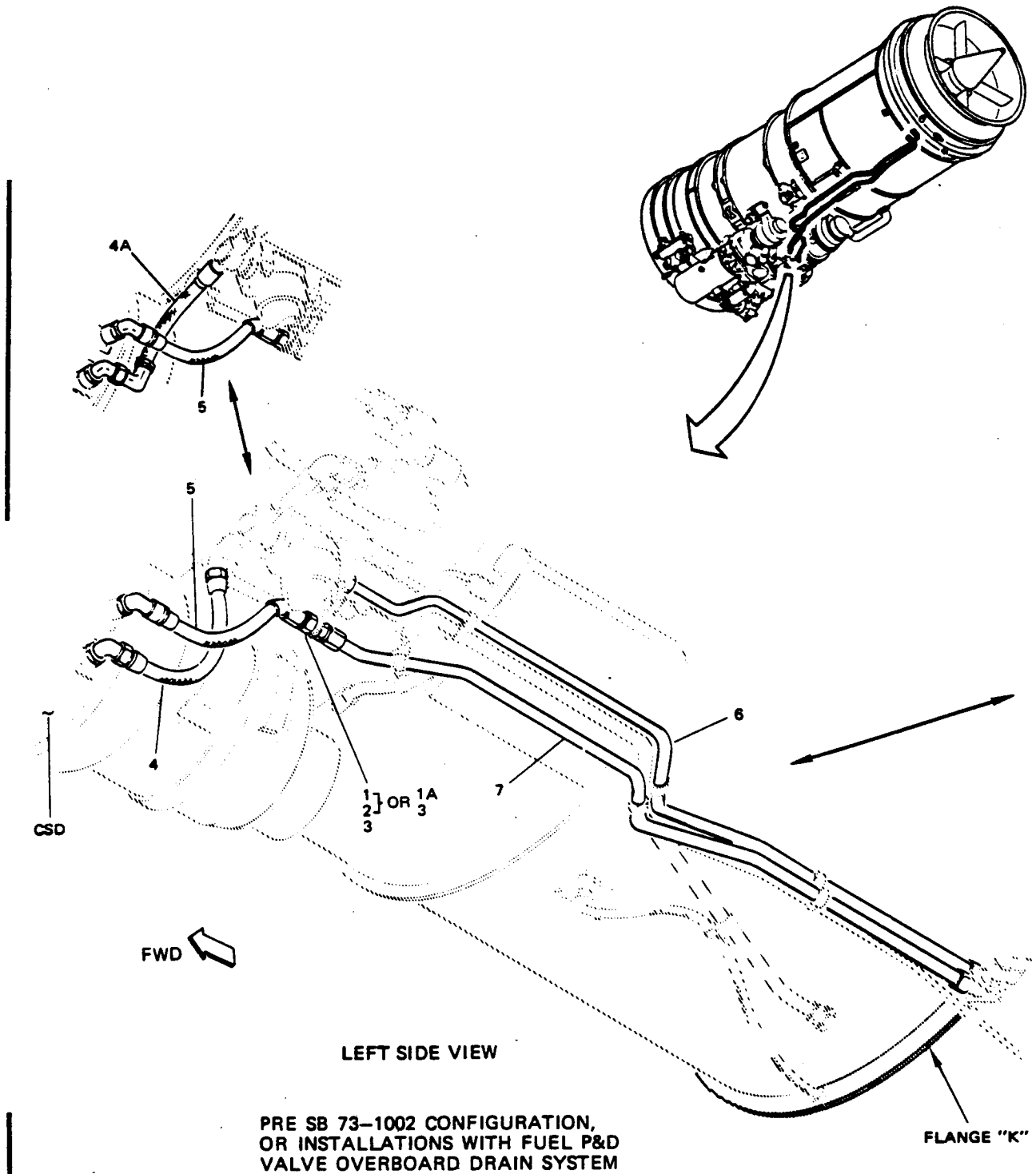
Aft Engine Drain Fittings Installation
Figure 551 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
AF. (1)	1	FITTING ASSEMBLY, ENGINE DRAIN	(PREASSEMBLED IN PAR. 3.P.) (ASSEMBLY 1)		RF	
	2	BRACKET	69-62249-2		1	
	2	BRACKET	69-52283-1		1	
	3	SCREW	BACS12CB3-8		2	
	4	WASHER	AN960-10L		2	
	5	NUT	BACN10JC3		2	
	(2)	6	FITTING ASSEMBLY, ENGINE DRAIN	(PREASSEMBLED IN PAR. 3.P.) (ASSEMBLY 2)		RF
		7	SCREW	BACS12CB3-7		2
		8	WASHER	AN960-10L		2
		9	NUT	BACN10JC3		2
	(3)	10	CLAMP	BACC10HS05		1
		11	SCREW	BACS12CB3-6		1
		12	NUT	BACN10JC3		1
	(4)	13	BRACKET ASSEMBLY	69-52291-1		1
		14	SCREW	BACS12CB3-7		4
		15	WASHER	AN960-10L		4
		16	NUT	BACN10JC3		4
	(5)	17	BRACKET	69-52283-2		1
		18	SCREW	BACS12CB3-6		2
				NAS1801-3-6		2
		19	NUT	NAS679A3W		2
		20	NUT	207865		2
		20A	JUMPER ASSY	BACJ40AD57-4		1
		20B	JUMPER ASSY	BACJ40AD57-6		1
		20C	WASHER	AN960C616L		4
		21	CLAMP ASSEMBLY	TA5000BH14AT		1
				TA5000BH16AT		1
		22	CLAMP	BACC10BH3RW		4
		23	SCREW	BACS12CB3-6		4
		24	NUT	BACN10JC3		4
		25	JUMPER	BACJ40AD45-9		1
26	SCREW	NAS1801-3-6		1		
27	WASHER	AN960C10L		1		
28	NUT	BACN10JC3		1		
29	BOLT	NAS6604-1		1		
30	WASHER	AN960C416L		2		
31	NUT	BACN10JC4		1		

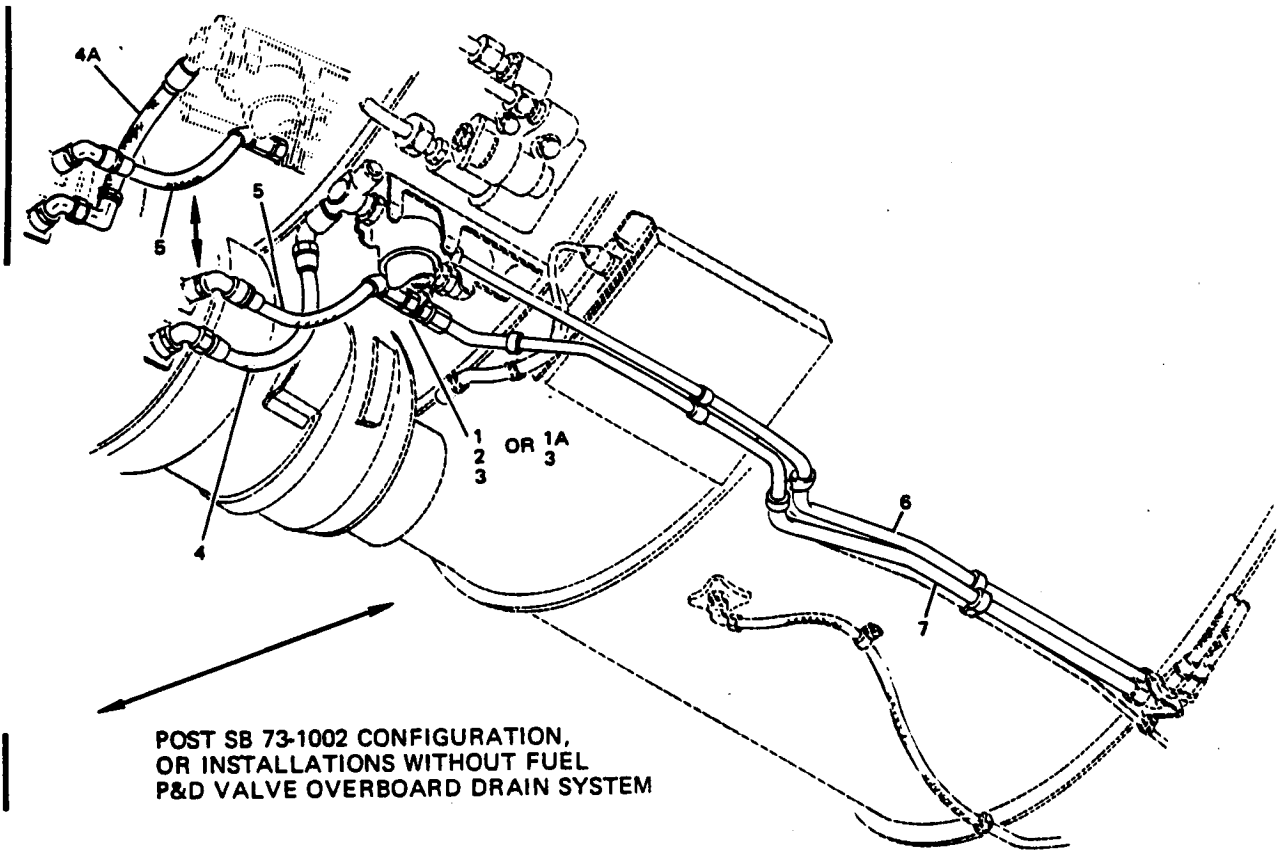
AG. Install CSD pressure system plumbing as follows: (Fig. 552)

- (1) Install union and washer (1, 2), or elbow (1A) and nut (3) on bracket attached to CSD oil filter assembly between flanges H and J on lower left side of engine.
- (2) Turn tee in CSD oil filter to horizontal position and attach to hose (4). Attach other end of hose to fitting in OUT port in CSD case and tighten hose end nuts. (65-55749-8 -- tighten to 240-360 lb-in.) Tighten jamnut on CSD OUT port to 90-110 lb-in. Tighten jamnut on tee.
- (3) Install hose (5) between IN port in CSD case and union (1) or elbow (1A). Tighten end nut of hose (5) at CSD OUT port. (65-55749-8 -- Tighten to 240-360 lb-in.) Tighten jamnut in CSD IN port fitting to 120-150 lb-in.
- (4) Install tube assembly (6) between union in OUT port of CSD filter and elbow on flange K (installed in par. AE.). Tighten tube end nuts.
- (5) Install tube assembly (7) between union (1) or elbow (1A) and elbow on flange K (installed in par. AE.). Tighten tube end nuts.

NOTE: Tube supporting clamps will be installed when P & D dump valve drain tube is installed in par. AH.



REF.
FIG
1120



CSD Pressure System Plumbing
Figure 552 (Sheet 2)

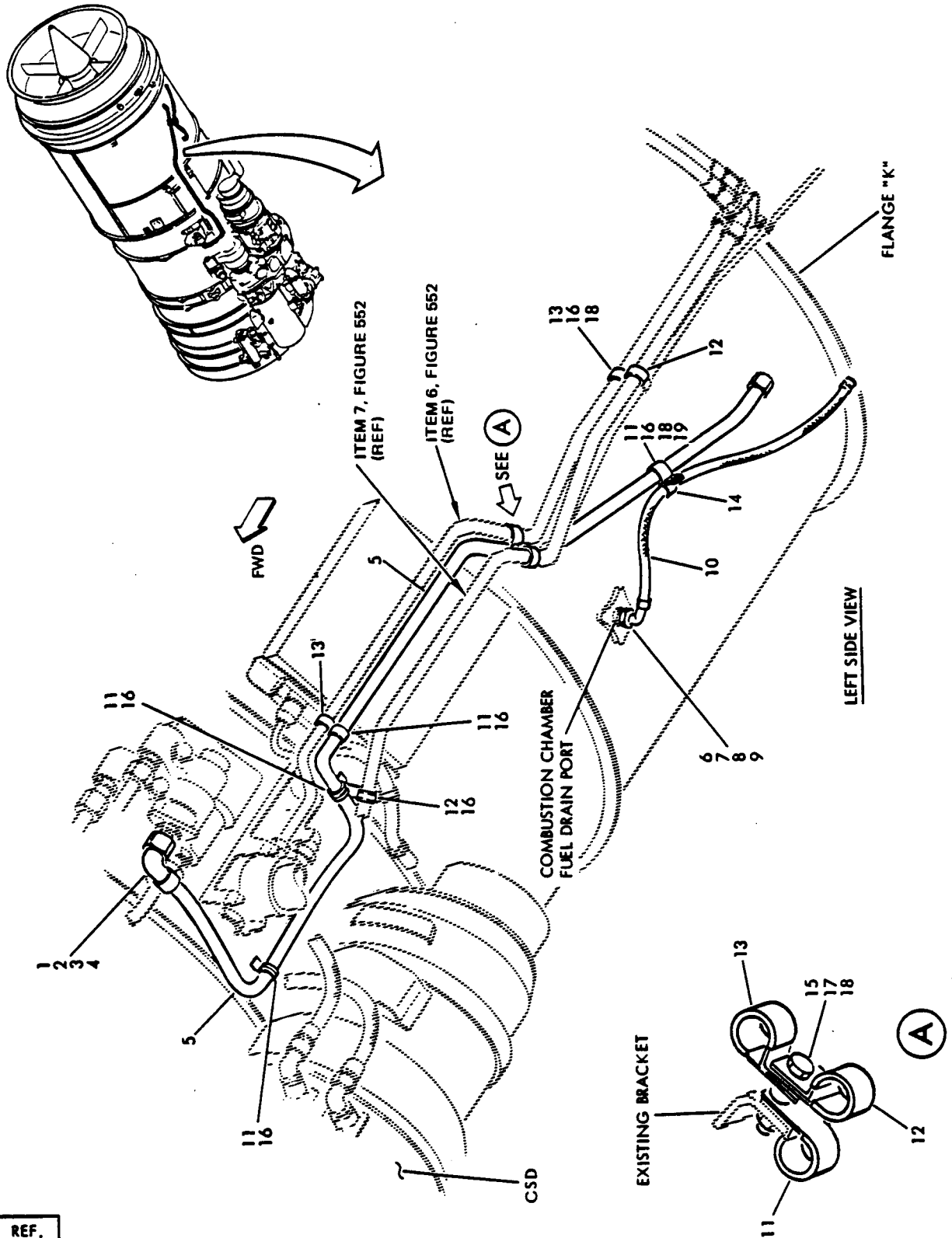
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AG. (1)	1	UNION	MS21924-10		1
	1A	ELBOW	MS21907-10		1
	2	WASHER	AN960-1416		1
	3	NUT	AN924-10		1
(2)	4	HOSE	BACH30AU08F0116		1
	4	HOSE	BACH5R0116DUF		1
	4A	HOSE	BACH30AU08F0116		1
(3)	5	HOSE	BACH30AV10F0114		1
	5	HOSE	BACH5S0074EVF		1
(4)	6	TUBE ASSEMBLY	65-55749-4		1
	6	TUBE ASSEMBLY	65-55749-6 (SB 73-1002)		1
(5)	7	TUBE ASSEMBLY	65-55749-5		1

AH. Install P & D drain tube and combustion chamber fuel drain tubes as follows (Fig. 553):

NOTE: Steps (1), (2), (3) not required when SB 73-1002 is incorporated.

- (1) Install jamnut (1) on elbow (2) and install elbow in P & D drain port with O-ring (3) and backup ring (4). Do not tighten jamnut.
- (2) Attach tube assembly (5) to elbow (2). Align tube assembly for clamping to CSD oil pressure system tubes, and tighten tube end nut. Tighten jamnut (1).
- (3) Attach aft end of tube assembly (5) to union in drain fitting assembly (installed in par. AF) on flange K.
- (4) Install jamnut (6) on elbow (7) and install elbow in combustion chamber fuel drain port FDI with O-ring (8) and backup ring (9). Do not tighten jamnut.
- (5) Install hose (10) on elbow (7). Align hose for clamping to P & D drain tube and tighten hose end nut. Tighten jamnut (6).
- (6) Attach aft end of hose (10) to drain fitting assembly (installed in par. AF) on flange K.
- (7) Ensure tube assembly (7, Fig. 552) is located directly under tube assembly (6, Fig. 522) with appropriate clearance from ignition exciter box.
- (8) Install clamps (11, 12, 13, and 14) and spacer (15) with screws (16 and 17) and nuts (18).

OVERHAUL MANUAL



REF.
FIG.
1119

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

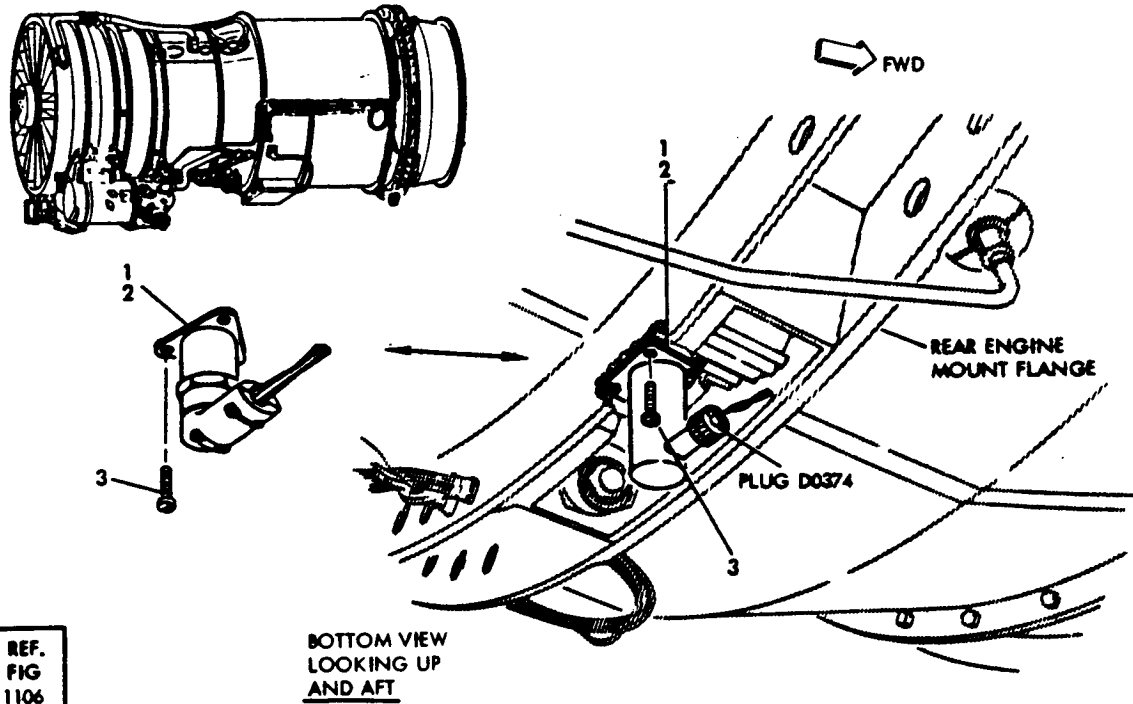
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AH.					
(1)	1	JAMNUT	AN6289-12		
*[1]	2	ELBOW, 90° BULKHEAD	MS21908-12		1
	3	O-RING	BACP11P-B12		1
	4	RING, Backup	MS28777-12		1
(2)	5	TUBE ASSEMBLY	65-55475-27		1
*[1]	5	TUBE ASSEMBLY	65-55475-11		1
(3)		NO PARTS REQUIRED			
(4)	6	JAMNUT	AN6289-6		1
	7	ELBOW, 90°	MS21926-6		1
	8	O-RING	BACP11P-B6		1
	9	RING, BACKUP	MS28777-6		1
(5)	10	HOSE	BACH30AS06W0201		1
	10	HOSE	BACH5P0184CCF (SB 73-1002)		1
(6)		NO PARTS REQUIRED			
(7)	11	CLAMP	BACC10HS12		5
	12	CLAMP	BACC10HS10		3
	13	CLAMP	BACC10HS08		3
	14	CLAMP	BACC10HS14		1
	15	SPACER	NAS43HT3-64		1
	16	SCREW	BACS12CB3-8		6
	17	SCREW	BACS12CB3-26		1
	18	NUT	NAS679A3W		AR
	19	SPACER	NAS43HT3-56 (SB 73-1002)		1

*[1] NOT REQUIRED WHEN SB 73-1002 INCORPORATED

AI. Install turbine vibration pickup as follows: (See figure 554.)

NOTE: Perform step (1) only if vibration pickup P/N 1784340-10 is used.

- (1) Attach vibration pickup (1) to adapter (2). Secure pickup to adapter with lockwire MS20995NC20.
- (2) Locate vibration pickup on vibration pickup pad between flanges of rear engine mount at bottom centerline of engine. Attach pickup with screws (3).
- (3) Install lockwire MS20995NC20 in a continuous double twist loop through all screws.
- (4) Attach electrical plug D0374 to connector on pickup and lockwire in place.



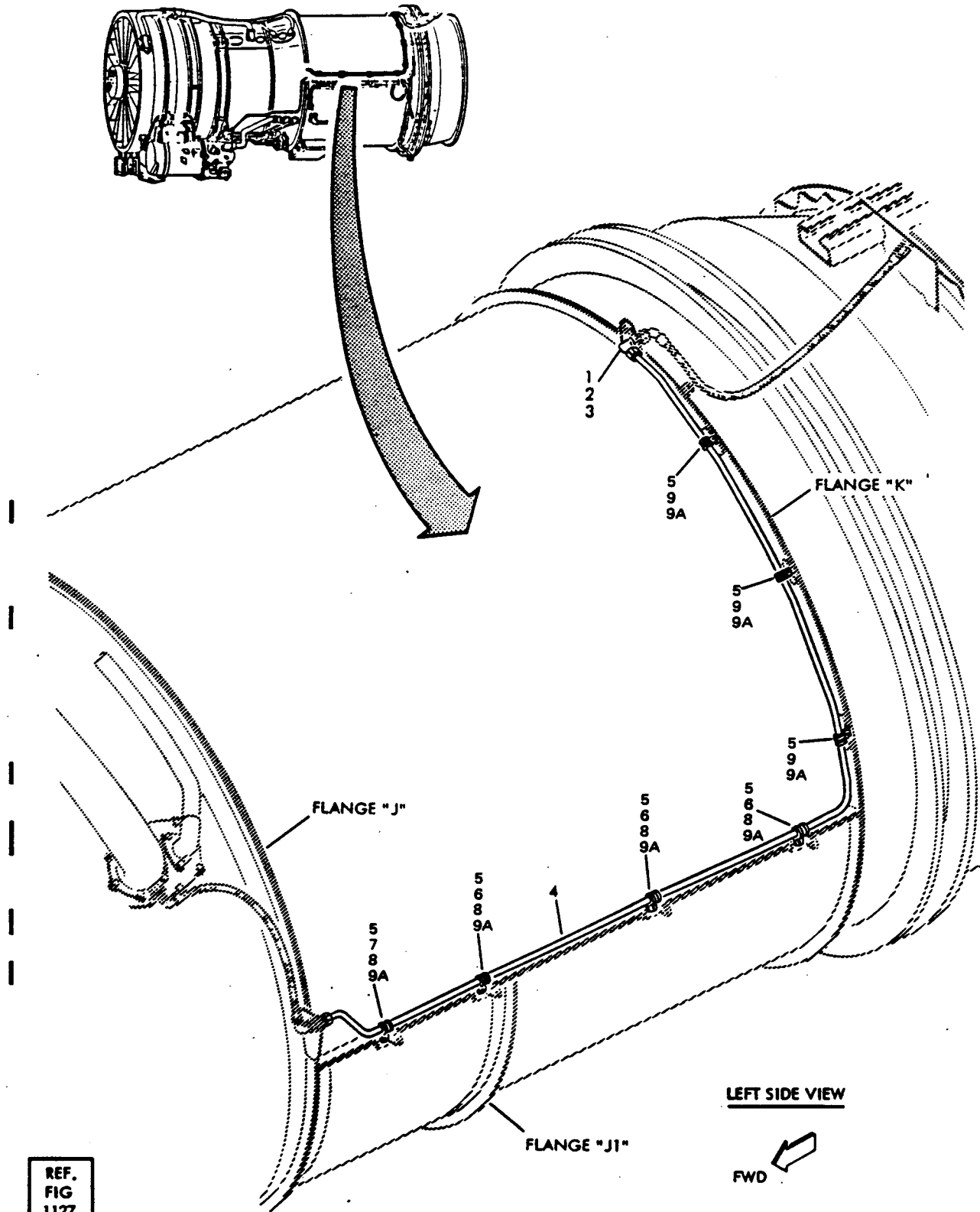
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AI.					
(1)	1	PICKUP, VIBRATION	4-125-0001		1
	1	PICKUP, VIBRATION	4-126-0001		1
	1	PICKUP, VIBRATION	WL-800		1
	1	PICKUP, VIBRATION	MODEL 750		1
	1	PICKUP, VIBRATION	1784340-10		1
	1	PICKUP, VIBRATION	MODEL 9145		1
	1	PICKUP, VIBRATION	WL-750		1
	1	PICKUP, VIBRATION	1784340-60		1
	1	PICKUP, VIBRATION	144-904-000-01		1
	1	PICKUP, VIBRATION	144-904-000-04		1
	1	PICKUP, VIBRATION	144-904-000-05		1
	2	ADAPTER	1510376-2 (USE WITH 1784340-10 ONLY)		2
(2)	3	SCREW	AN500AD6-8		4
	3	SCREW	AN500AD8-7 (USE WITH 1784340-10 ONLY)		3
(3)		NO PARTS REQUIRED			
(4)		NO PARTS REQUIRED			

OVERHAUL MANUAL

AJ. Install hydraulic tubes as follows (Fig. 555):

- (1) Install elbow (1), washer (2), and nut (3) on existing bracket at top of flange "K."
- (2) Install tube assembly (4) between elbow (1) and existing union (installed in par. N) on left side of flange "J."
- (3) Install clamps (5) and spacers (6 and 7) with washers (9A) and screws (8 and 9).
- (4) Tighten end fittings of tube assembly (4). Tighten nut (3).
- (5) Deleted
- (6) Cap open end of tube.

OVERHAUL MANUAL



REF.
FIG
1127

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AJ. (1)	1	ELBOW	MS21908-4		1
	2	WASHER	AN960-716		1
	3	NUT	AN924-4		1
(2)	4	TUBE ASSEMBLY	65-58145-11		1
	4	TUBE ASSEMBLY	65-58145-3		1
(3)	5	CLAMP	BACC10HSO4		7
	6	SPACER	NAS43HT3-20		3
	7	SPACER	NAS43HT3-32		1
	8	SCREW	BACS12CB3-(-)*[1]		4
	9	SCREW	BACS12CB3-(-)*[1]		3
	9A	WASHER	AN960D10		7
(4)		NO PARTS REQUIRED			
(5)	10	DELETED			
(6)		NO PARTS REQUIRED			

*[1] LENGTH AS REQUIRED

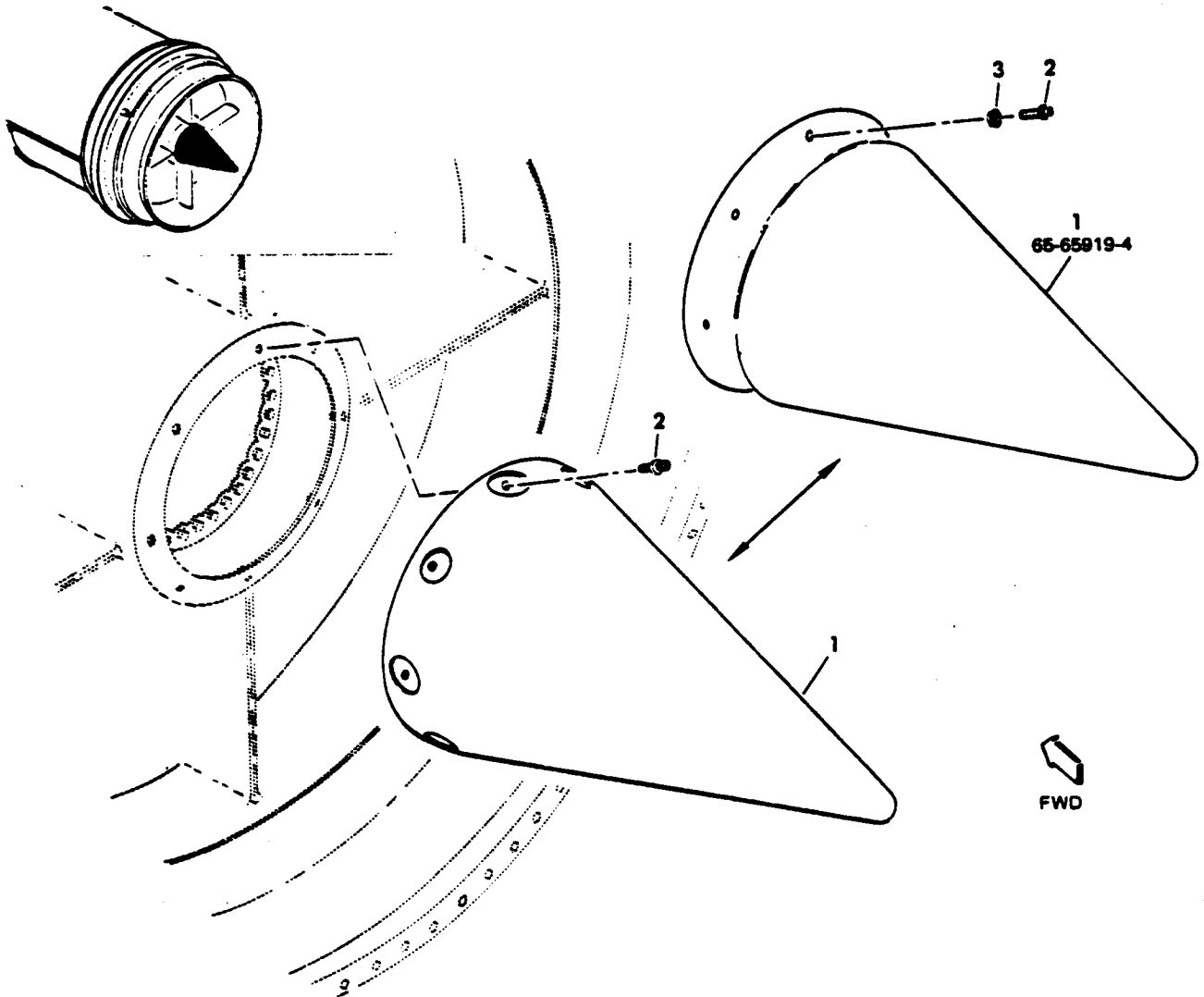
AK. Install tail plug as follows: (See Fig. 556)

(1) Locate tail plug (1) on engine plug support. Attach with eight bolts (2) and washers (3).

(2) Tighten bolts (2) as follows:

Dry Bolt - 65 to 79 lb-in
Lubed Bolt - 62 to 76 lb-in

NOTE: Bolts may be lubed using compounds such as Ease-Off 990, Bostik, Never-Seez, MIL-C-11796, or MIL-G-23287.



REF.
FIG
1108

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AK. (1)	1	PLUG, TAIL	65-65919-4		1
	1	PLUG, TAIL	65-27808-14 (OPT)		1
	1	PLUG, TAIL	65-27808-800 (OPT)		1
	2	BOLT	MS9034-10 (U/O 65-65919-4)		8
	2	BOLT	MS9034-08 (U/O 65- 27808-14,-800)		8
	3	WASHER	AN960C416		8

AM. Install tailpipe extension and shroud assembly as follows (Fig. 558).

CAUTION: WEAR CLEAN, WHITE, COTTON GLOVES WHEN HANDLING TITANIUM PARTS. ANY SALT DEPOSIT, EVEN FINGERPRINTS, ON UNPREPARED SURFACE CAN CAUSE TITANIUM ALLOY TO CORRODE WHEN SUBJECTED TO HIGH TEMPERATURES.

(1) Install pin (2) at top of tailpipe assembly (1) using washer (3) and nut (4).

(2) Install clamps (5) on tailpipe assembly (1) with rivets (6).

NOTE: Step (2) applies to installations using the Kidde fire detector system.

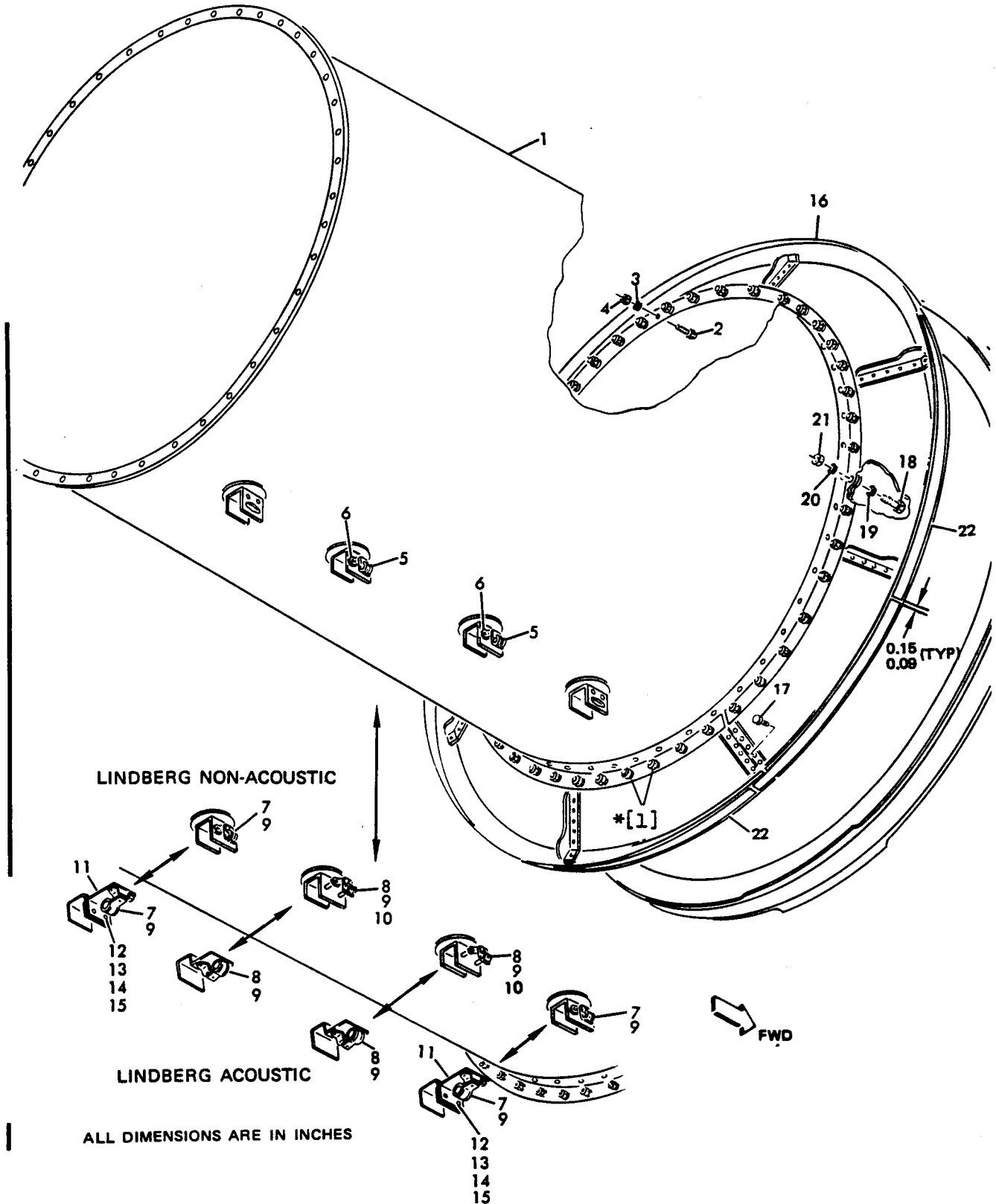
(3) If Lindberg fire detector system is used with a non-acoustic tailpipe, install clamps (7 and 8) with rivets (9) and spacers (10).

(4) If Lindberg fire detector system is used with an acoustic tailpipe, install brackets (11) with screws (12), spacers (13), washers (14), and nuts (15). Install clamps (7 and 8) with rivets (9).

(5) Assemble two halves of shroud assembly (16) on tailpipe with bolts (17).

(6) Install tailpipe assembly (1) and shroud assembly (16) on engine aft flange. Secure tailpipe and shroud assemblies with bolts (18), washers (19 and 20) and nuts (21). Install washer (19) under bolt head and washer (20) under nut.

(7) Locate rub strips (22) around periphery of shroud assy (16), except part of top segment. Space rub strips within 0.09-0.15 inch apart. Bond etched side of rub strips per 20-50-12 type 44, using BMS 5-44, class B sealant. Bond to be 0.010-0.060 inch thick.



Tailpipe Extension and Shroud Assembly Installation
 Figure 558 (Sheet 1)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
5. AM. (1)	1	TAILPIPE ASSY (ACOUSTIC)	65-82788-1			
	1	TAILPIPE ASSY	65-82788-6		1	
	1	TAILPIPE ASSY (ACOUSTIC)	65-82788-16		1	
	1	TAILPIPE ASSY	65-70190-1		1	
	2	PIN	65-70197-4		1	
	3	WASHER	AN960C10L		1	
	4	NUT	BACN10GW3A		1	
	(2)	5	CLAMP	TA12100004		2
		5	CLAMP	TA5000BH8AT		2
		6	RIVET	MS20615-6-		4
	(3)	7	CLAMP	TA5000BH14AT		2
		8	CLAMP	TA12100004		2
		8	CLAMP	TA5000BH8AT		2
		9	RIVET	MS20615-6-		8
		10	SPACER	NAS1056E6-031		4
(4)	11	BRACKET	65-84112-6		2	
	12	SCREW	BACS12CB3-8		4	
	13	SPACER	NAS43HT3-6		4	
	14	WASHER	AN960-C10		4	
	15	NUT	BACN10JC3		4	
(5)	16	SHROUD ASSY	65-70191-1		1	
	16	SHROUD ASSY	65-70191-34		1	
	17	BOLT	BACB30LK3U2		21	
(6)	18	BOLT *[1]	BACB30LE5U6		42	
	19	WASHER	BACW10BP5ACU		42	
	20	WASHER	AN960516L		42	
	21	NUT	BACN10HR5C		42	
(7)	22	RUB STRIP	65-70191-6		6	

*[1] Then grounding straps are used with Kidde fire detector, replace two flange bolts with BACB30LE5U8 at locations shown

AN. Deleted

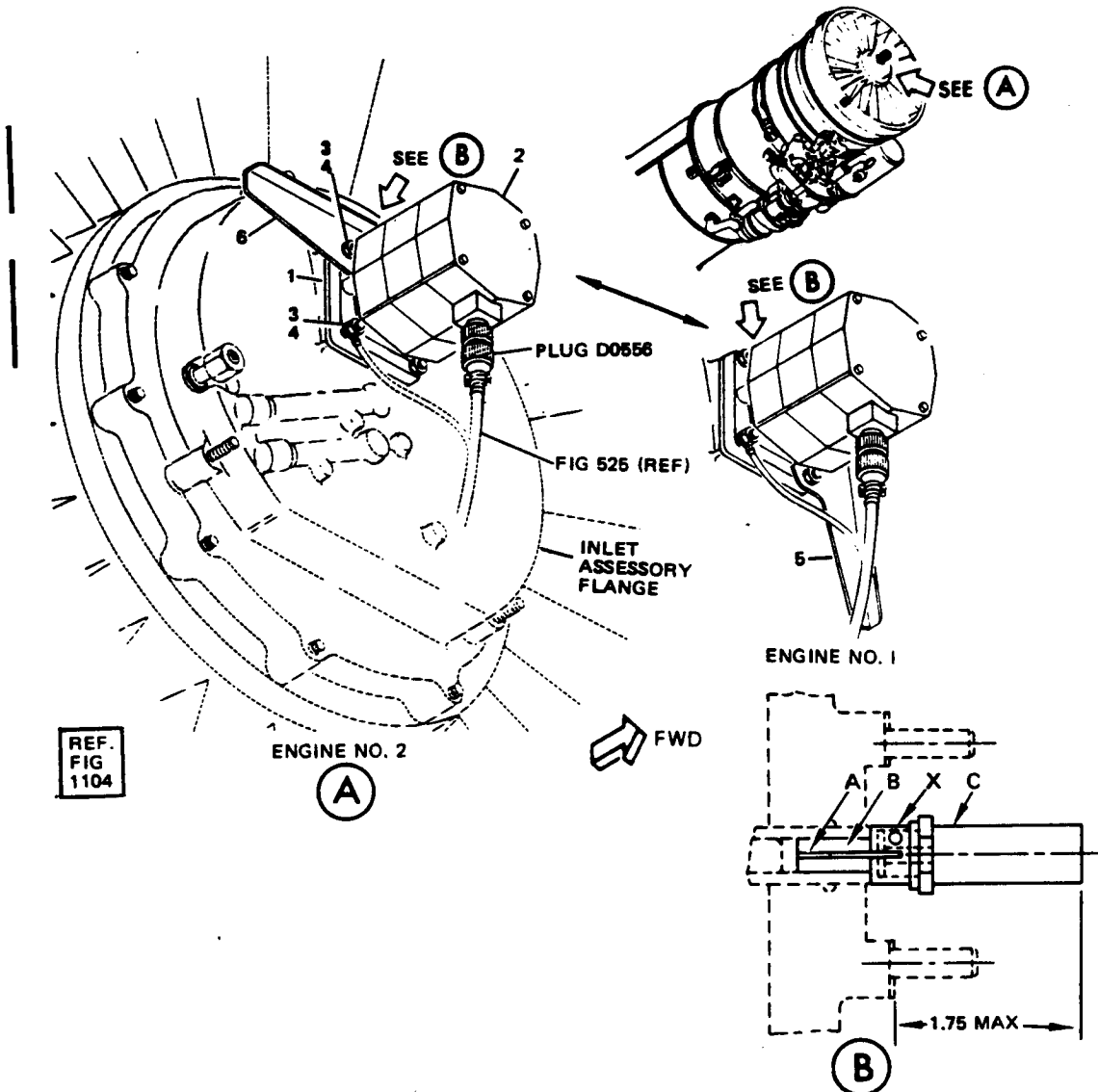
AO. Install N1 tachometer generator as follows (Fig. 560):

- (1) Place gasket (1) over studs on N1 tachometer mounting pad.
- (2) Tachometer Generator (2)
 - (a) B5069110302-0 or 2CM9ABY7 -- Coat drive shaft of tachometer generator with grease MIL-G-81322 per 20-50-07.
 - (b) 2CM15AAB1 -- Before installation perform following (SEE **(B)**):
 - 1) Visually inspect surface x of rotor part B and C for any particles or other contamination that might interfere with alignment.
 - 2) Clean if necessary and relubricate surfaces x with lubricant per MIL-A-907D.
 - 3) Insert square end of rotor assy into engine shaft and tighten part C finger-tight.
 - 4) Tap end of rotor C gently with brass or plastic hammer to assure seating with engine shaft, and tighten rotor C to 40 lb-in.
- (3) Install tachometer generator and dome interference bracket (5 or 6), if required, with washers (3) and nuts (4). Tighten nut (4) to 65-70 lb-in. Place ground wire of wire bundle W196 beneath washer on one stud (reference electrical installation, Fig. 525).

NOTE: The interference brackets, when installed on the correct engines, prevent improper interchange of nose domes.

- (4) Attach plug D0556 to electrical connector on tachometer generator and lockwire in place.

NOTE: Make electrical bond between interference brackets and accessory drive housing. Maximum allowable resistance is 0.0025 ohm (2.5 milliohms).



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
A0. (1)	1	GASKET	AN4045-1		1
(2)	2	TACHOMETER GENERATOR	B5069110302-0		1
	2	TACHOMETER GENERATOR	2CM9ABY7		1
	2	TACHOMETER GENERATOR	2CM15AAB1		1
(3)	3	WASHER	AN960PD416L		4
	4	NUT	NAS679A4W		4
	5	INTERFERENCE BRACKET	65C13050-1		1
	6	INTERFERENCE BRACKET	65C13050-2		1
(4)		No parts required			

N1 Tachometer Generator Installation
Figure 560

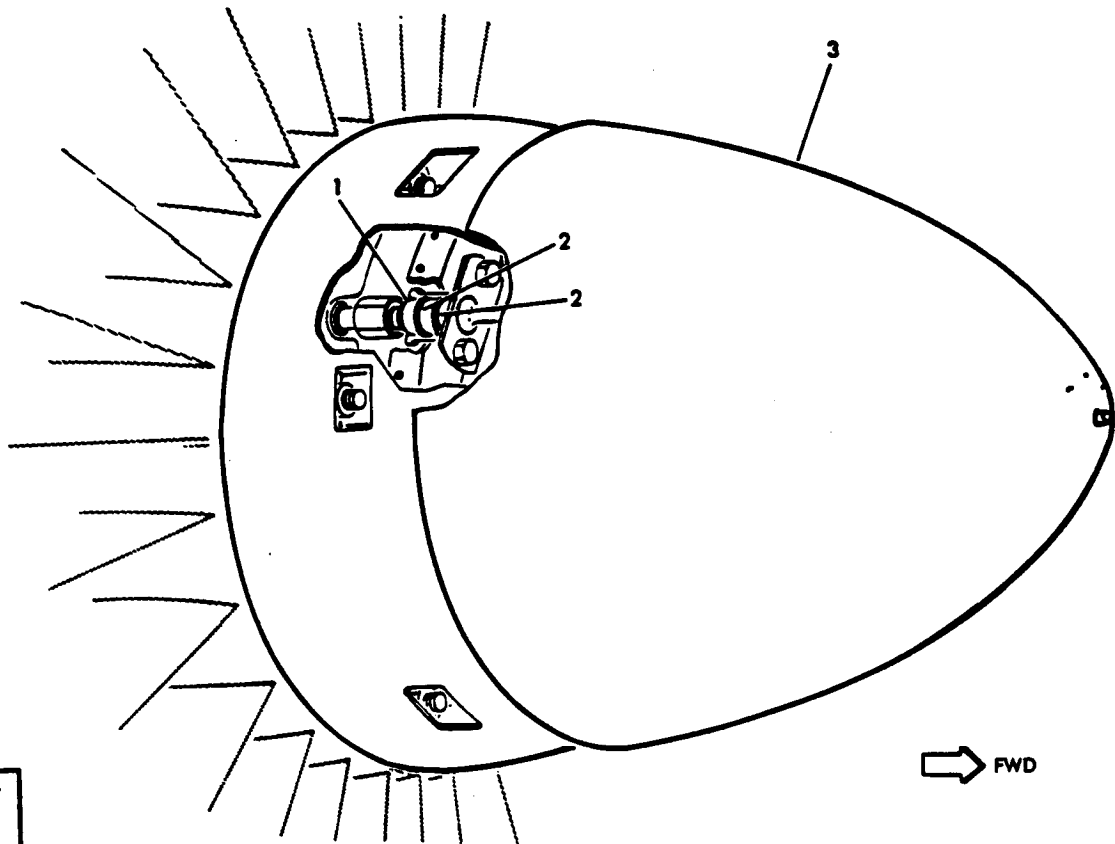
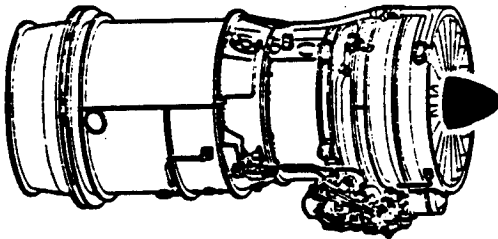
AP. Install nose dome assembly as follows: (Fig. 561)

CAUTION: LEFT AND RIGHT HAND NOSE DOME ASSEMBLIES ARE NOT INTERCHANGEABLE. ENSURE THAT INTERFERENCE BRACKETS ARE INSTALLED UNDER N1 TACHOMETER (PER FIG. 560) TO PREVENT IMPROPER NOSE DOME INSTALLATION.

- (1) Apply grease, Specification MIL-G-4343 to O-rings (2) and install with PT2 coupler (1).

CAUTION: TO AVOID FALSE EPR READINGS THE PT2 TUBE MUST PROJECT 0.080 TO 0.120 INCH BEYOND THE NOSE DOME.

- (2) Apply grease, Specification MIL-G-4343 to engine front accessory drive housing O-ring and position nose dome assembly (3) on engine front accessory drive housing taking care to align mounting studs and Pt2 probe connection.
- (3) Tighten four captive nuts in dome assembly in diametrically opposite pairs 100 to 140 pound-inches.



REF.
FIG
1102
1143

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
AP. (1)	1	COUPLER, PT2	69-18713-800		1
	2	O-RING	NAS1611-113 BACP11J113 (PREFD)		2
(2)	3	DOMES, Nose	65-19684-2 or 65-85369 *[1]		1
(3)		No parts required			

*[1] 65-85369-() IS USED WITH NOSE COWL 65-85378-() (REF FIG. 1143),
65-19684-2 IS USED WITH NOSE COWL 65-60529-() (REF FIG. 1102).

737

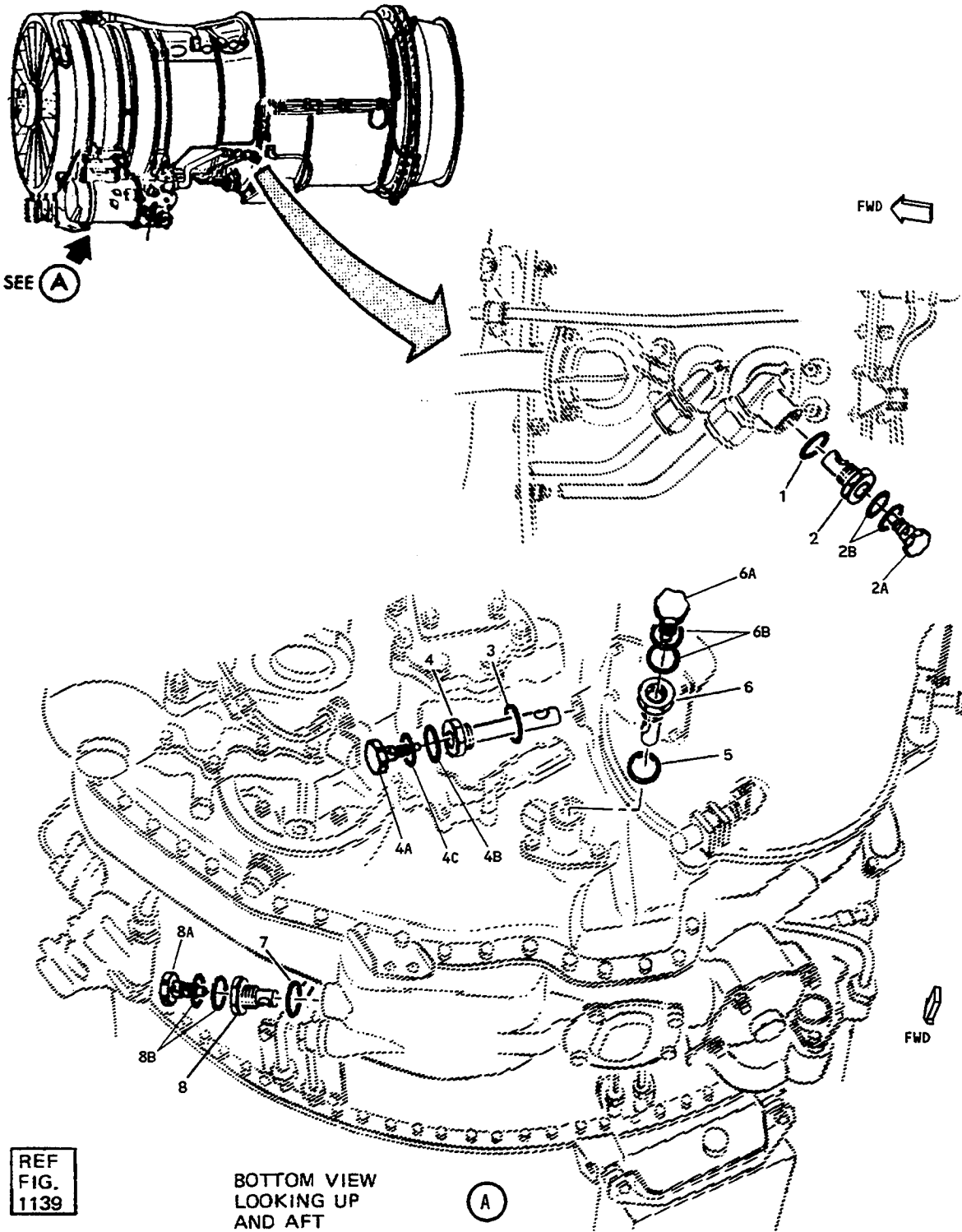
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

AQ. Install chip detectors as follows: (See figure 561A.)

CAUTION: ENSURE THAT O-RING IS PROPERLY INSTALLED IN TURBOMAG PLUGS (2, 4, 6, 8) AND PACKING (1, 3, 5, 7) IS PROPERLY USED ON INSTALLATION OF TURBOMAG PLUGS. ENGINE OIL LOSS MAY RESULT FROM MISSING OR IMPROPERLY INSTALLED O-RINGS.

- (1) Remove P & WA plug MS9015-05 and packing 385990. Install packing (1) and plug (2). Tighten plug (2) to 8-40 lb-in. Lockwire plug (2) per 20-50-02.
- (2) Remove P & WA plug MS9015-07 and packing 382061. Install packing (3) and plug (4). Tighten plug (4) to 20-60 lb-in. Lockwire plug per 20-50-02.
- (3) Remove P & WA plug MS 9015-05 and packing 385990. Install packing (5) and plug (6). Tighten plug (6) to 8-40 lb-in. Lockwire plug per 20-50-02.
- (4) Remove P & WA plug MS9015-04 and packing 379643. Install packing (7) and plug (8). Tighten plug (8) to 30-108 lb-in. Lockwire plug per 20-50-02.



REF
 FIG.
 1139

BOTTOM VIEW
 LOOKING UP
 AND AFT

(A)

Chip Detector Installation
 Figure 561A (Sheet 1)

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET

OVERHAUL MANUAL

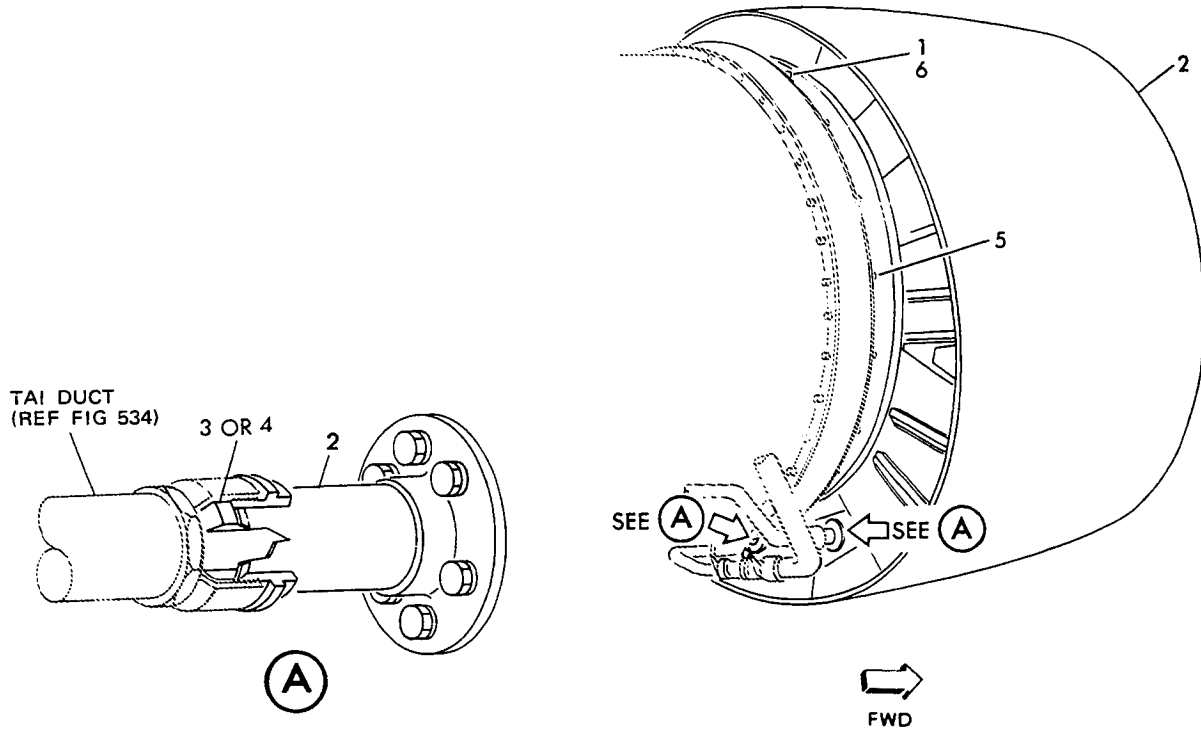
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NUMBER	QTY
AQ (1)	1	PACKING, O-RING	BACP11P-B5A		1
	2	TURBOMAG PLUG, V97484	B7670		1
(2)	2A	MAGNETIC PLUG, V97484	B6011		1
	2B	PACKING O-RING	M8324811-008		2
	3	PACKING, O-RING	BACP11P-B6A		1
	4	TURBOMAG PLUG, V97484	B6002		1
(3)	4A	MAGNETIC PLUG, V97484	B6035		1
	4B	PACKING O-RING	M83248/1-008z		1
	4C	PACKING O-RING	M83248/1-001		1
	5	PACKING, O-RING	BACP11P-B5A		1
(4)	6	TURBOMAG PLUG, V97484	B7670		1
	6A	MAGNETIC PLUG, V97484	B6011		1
	6B	PACKING O-RING	M83248/1-008		2
	7	PACKING, O-RING	BACP11P-B8A		1
(4)	8	TURBOMAG PLUG, V97484	B6003		1
	8A	MAGNETIC PLUG, V97484	B6032		1
	8B	PACKING O-RING	M83248/1-012		2

Chip Detector Installation
Figure 561A (Sheet 2)

6. Final Buildup of Engine No. 1 Only

A. Install nose cowl on engine No. 1 as follows (Fig. 562):

- (1) For 65-60529 series nose cowl only, install 20 bolts (1) loosely in engine flange in positions corresponding to slotted holes in nose cowl attaching ring.
- (2) Install 65-60529 series nose cowl (2) over bolt heads and rotate so bolts butt against narrowed end of slots. If 65-85378 series nose cowl is used, position on engine flange and install 23 bolts (1) and 23 washers (6).
- (3) Verify that seal ring (3) in TAI duct and seal ring (4) in vortex dissipator duct are in proper position, and loosely connect ducts to nose cowl (Fig. 534).
- (4) For 65-60529 nose cowl, install three bolts (5) through three unslotted holes in nose cowl attaching ring.
- (5) Tighten bolts (1 and 5) to 125-155 pound-inches.
- (6) Tighten and lockwire TAI fitting and vortex dissipator fitting.



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
6.A. (1)	1	BOLT	MS9148-08 (used with 65-85378 nose cowl) (Fig. 1143)		23
(1)	1	BOLT	MS9148-08 (used with 65-60529 nose cowl) (Fig. 1121)		20
(2)	2	NOSE COWL ASSY	Refer to Fig. 1121 and 1143 for part No.		1
(3)	3	SEAL RING, TAI Duct	NAS595-20E		Ref
	4	SEAL RING, Vortex dissipator duct	NAS595-24E		Ref
(4)	5	BOLT	MS9148-08 (used with 65-60529 nose cowl) (Fig. 1121)		3
(2)	6	WASHER	AN960D516 (used with 65-85378 nose cowl) (Fig. 1143)		23
(5)		No parts required			
(6)		No parts required			

Nose Cowl Installation - Engine No. 1
Figure 562

CAUTION: THRUST REVERSER WEIGHS APPROXIMATELY 390 LBS. SUPPORT ENTIRE ASSY DURING INSTALLATION USING SLING F80110.

B. Install thrust reverser (Engine No. 1) (Fig. 563).

CAUTION: BOLT SUBSTITUTION IS NOT PERMITTED UNLESS SPECIFIED.

- (1) Viewing toward the engine tailpipe, position thrust reverser (1) on the engine tailpipe flange with index pin (5A) at location shown in Fig. 562A. Install bolts (2) with heads forward and washers (3) under boltheads. Install nuts, washers (5, 4). Substitution of AN960C516L for item (4) is permitted. One additional washer also may be used.
- (2) Install brackets (6) on forward face of tailpipe flange (Fig. 562A) using bolts, washers, nuts (7, 8, 9). Omit bolts at locations shown.
- (3) Install push-pull control assembly (10) (Teleflex, Fig. 563).

NOTE: Control and attaching parts are components of push-pull cable system.

- (a) Check cable for lubrication. If grease is required, apply Dow Corning 33 (light grade).
- (b) Install forward end of cable of control thru forward fairing and install (11, 12).
- (c) Install aft end of cable of control thru aft fairing and install (13, 14).

CAUTION: USE CARE TO PREVENT CONTAMINATION AND KINKING WHEN INSERTING CABLE THRU SHROUD BULKHEAD.

- (d) Position cable on tailpipe extension using parts (15 thru 20). Do not tighten.
- (e) Loosen nut at B (2 places) and rotate conduit to obtain best possible position and fit. Tighten nut and install lockwire. Tighten all bolts and install lockwire. See Fig. 562B for lockwire installation at bulkhead.
- (f) Attach push-pull assembly to bellcrank thru access door using parts (21, 22, 23).

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET

OVERHAUL MANUAL

- (4) Install push-pull control assembly (10) (Controlex, Fig. 563, 562B).

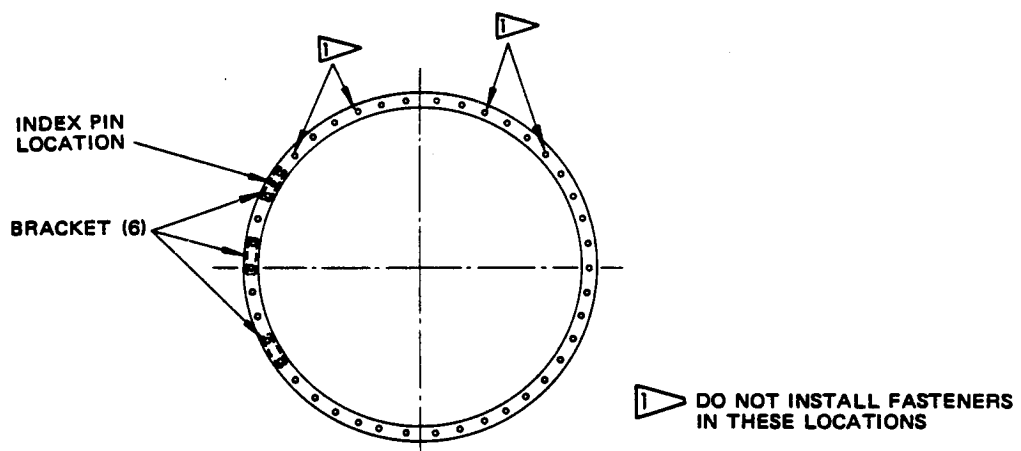
NOTE: Control and attaching parts are components of the push-pull cable system.

CAUTION: THIS ASSY CONTAINS A FLAT STEEL CENTER CORE WHICH BENDS EASILY IN THE FLAT PLANE BUT RESISTS BENDING IN THE EDGEWISE DIRECTION. DO NOT BEND BY FORCE TO ANY POSITION OR DAMAGE WILL RESULT.

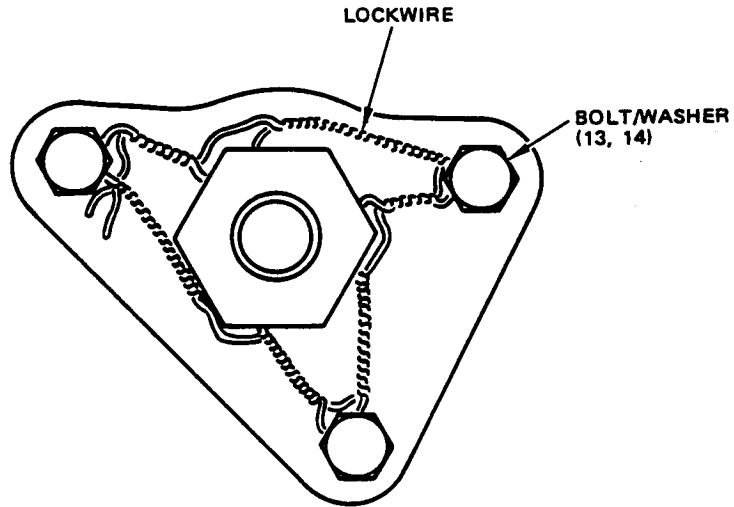
- (a) Install forward end of cable of control (10) thru forward fairing and install (11, 12).
- (b) Attach clamps (15) using (19A, 20). Do not tighten.

CAUTION: CABLE ASSY TELESCOPE ROD MUST BE CENTERED IN SWIVEL WHEN IN UNATTACHED POSITION OR DAMAGE WILL RESULT.

- (c) Insert aft telescopic unit thru aft fairing and install (13, 14). Use four washers at forward side (under head of union) and two washers at aft side (under union nut). Do not tighten bolts (13). Shifting of washers to either side of mounting plate is permitted for adjustment. However, a total of six washers must be used. Tighten union nut.
- (d) Attach brackets marked OUT and LH, or (18A) in place of bracket marked LH for 65-55835-13 and SB 78-1041 configurations, to tailpipe extension (with markings visible) using (19, 19B, 20). Do not tighten.



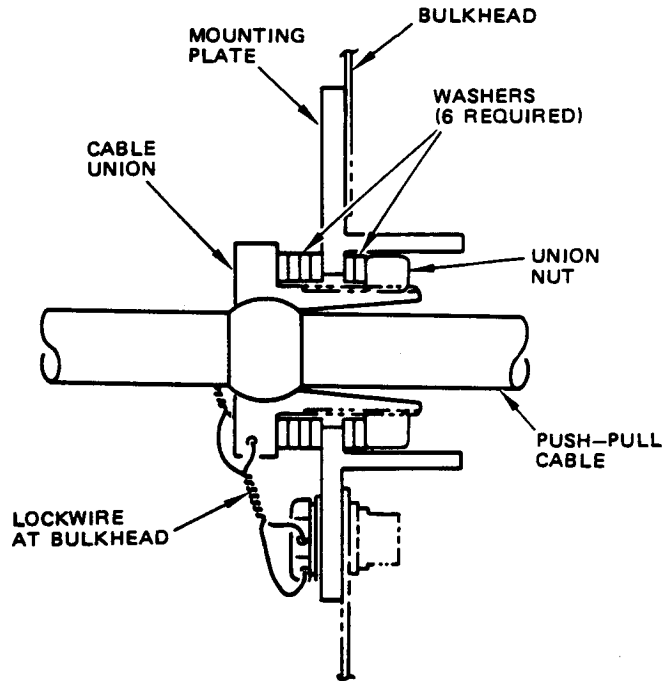
- (e) Provide sufficient loop to cable between connection at aft fairing (Ref step 4a) and bracket marked LH (Ref step 4d) to insure that no preload exists on the telescope rod when it is installed in thrust reverser lever.
 - (f) Adjust cable in brackets to obtain best possible position and fit. Maintain minimum clearance with tailpipe of 0.12 inch by adjusting cable thru clamps. After position is established, tighten all bolts and install lockwire (Fig. 562B).
 - (g) Attach push-pull assembly to bellcrank thru access door using parts (21, 22, 23).
- (5) Attach tubes at bulkhead fittings. Install tubes (24, 25) with clamps (26, 27) and fasteners (28, 29) at three places.



TELEFLEX INSTL
LOCKWIRE END NUT, JAM NUT,
& BOLTS

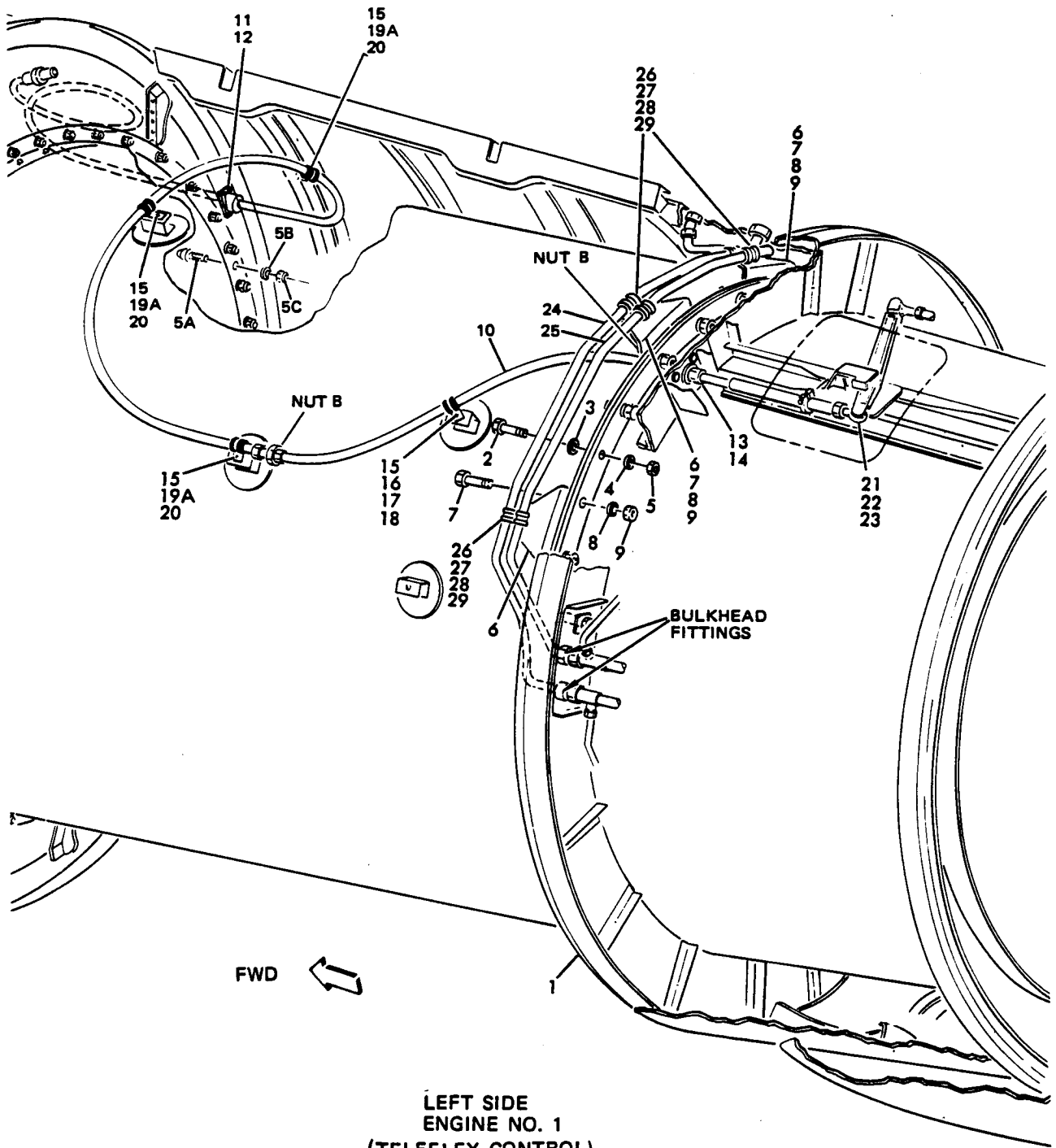
CONTROLEX INSTL
LOCKWIRE UNION & BOLTS

LOCKWIRE DETAILS



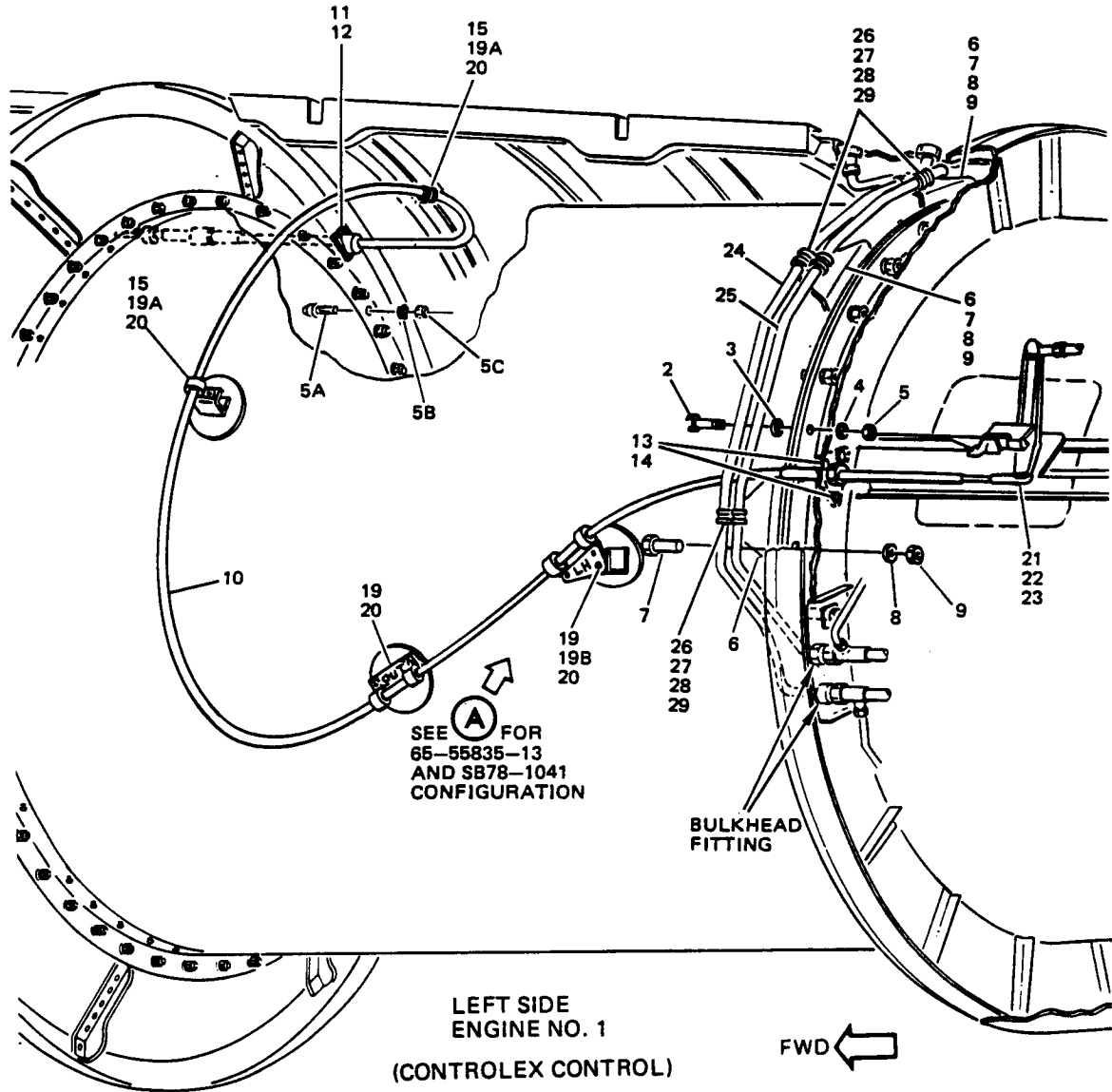
SECTION - AFT TELESCOPIC END OF
CONTROLEX CABLE

Push-Pull Control Installation Details
Figure 562B

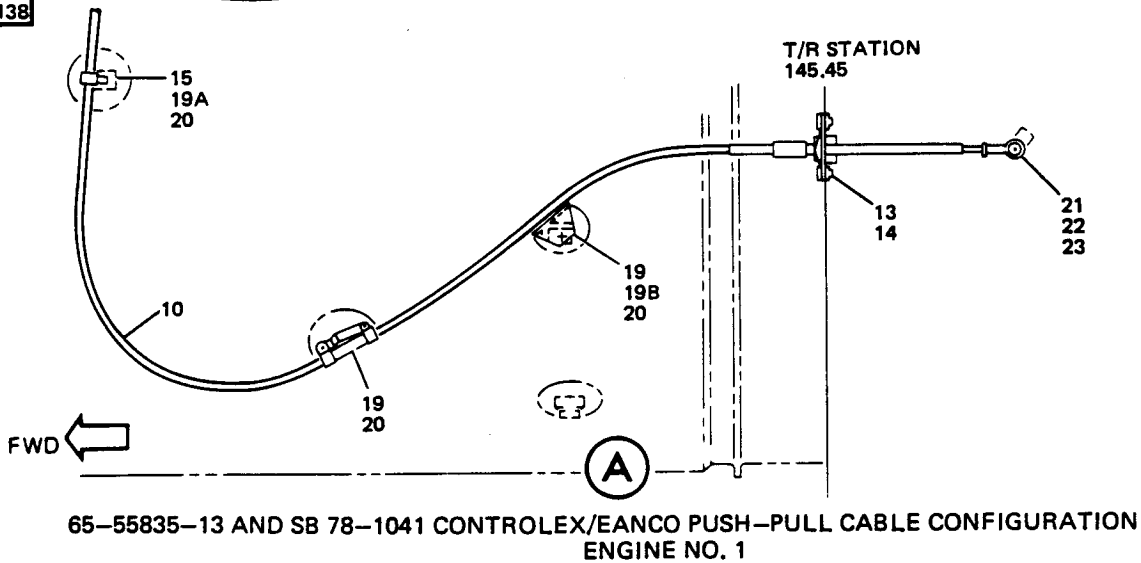


LEFT SIDE
ENGINE NO. 1
(TELEFLEX CONTROL)

REF
FIG.
1138



REF
 FIG
 1138



Thrust Reverser Installation - Engine No. 1
 Figure 563 (Sheet 2)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
6.B. (1)	1	THRUST REVERSER	10-61813-7		1	
	1	THRUST REVERSER	10-61813-27		1	
	1	THRUST REVERSER	10-61813-33		1	
	1	THRUST REVERSER	10-61813-57		1	
	2	BOLT	BACB30LE5U6		32	
	3	WASHER	BACW10BP5ACU		32	
	4	WASHER	AN960C516		32	
	5	NUT	BACN10HR5C		32	
	5A	PIN, INDEX	65-70197-4		1	
	5B	WASHER	AN960C10L		1	
	5C	NUT	BACN10GW3A		1	
	(2)	6	BRACKET ASSY	69-49854-1		3
		6	BRACKET ASSY	69-42854-3		3
		7	BOLT	BACB30LE5U6		6
		8	WASHER	AN960C516L		6
		9	NUT	BACN10HR5C		6
	(3,4)	10	CONTROL ASSY *[2]	25604		1
		10	CONTROL ASSY *[2]	C81662-2		1
		10	CONTROL ASSY *[2]	C81662-3		1
		11	BOLT	BACB30LM4U3		4
		12	WASHER	AN960C416L		4
		13	BOLT	BACB30LM4HU2		3
		14	WASHER	AN960C416L		3
		15	CLAMP *[3]	BACC10HS07		4
		15	CLAMP *[4]	BACC10HS07		2
		16	BOLT	BACB30LM3U2		1
		17	WASHER	AN960-10L		1
		18	NUT	BACN10JC3C		1
		18A	DEFLECTOR BRACKET	69-77002-1		2
		19	BOLT	BACB30LM3U3		3
		19A	BOLT *[3]	BACB30LM3U2		3
	19A	BOLT *[4]	BACB30LM3U2		2	
19B	NUT	BACN10JC3		2		
20	WASHER	AN960C10L		1		
21	BOLT	BACB30LJ4U10		1		
22	WASHER	AN960416L		1		
23	NUT	BACN10JC4C		1		
(5)	24	TUBE ASSY *[1]	65-46858-135		1	
	24	TUBE ASSY *[1]	65-46858-1135 (OPT)		1	
	24	TUBE ASSY	65C21843-1		1	
	25	TUBE ASSY *[1]	65-46858-139		1	
	25	TUBE ASSY *[1]	65-46858-1139 (OPT)		1	
	25	TUBE ASSY	65C21843-3		1	

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
6.B. (5)	26	CLAMP	BACC10HS06		3
	26	CLAMP	BACC10HS26		3
	27	CLAMP	BACC10HS08		3
	27	CLAMP	BACC10HS28		3
	28	BOLT	BACB30LM4U3		3
	29	NUT	BACN10JC3C		3

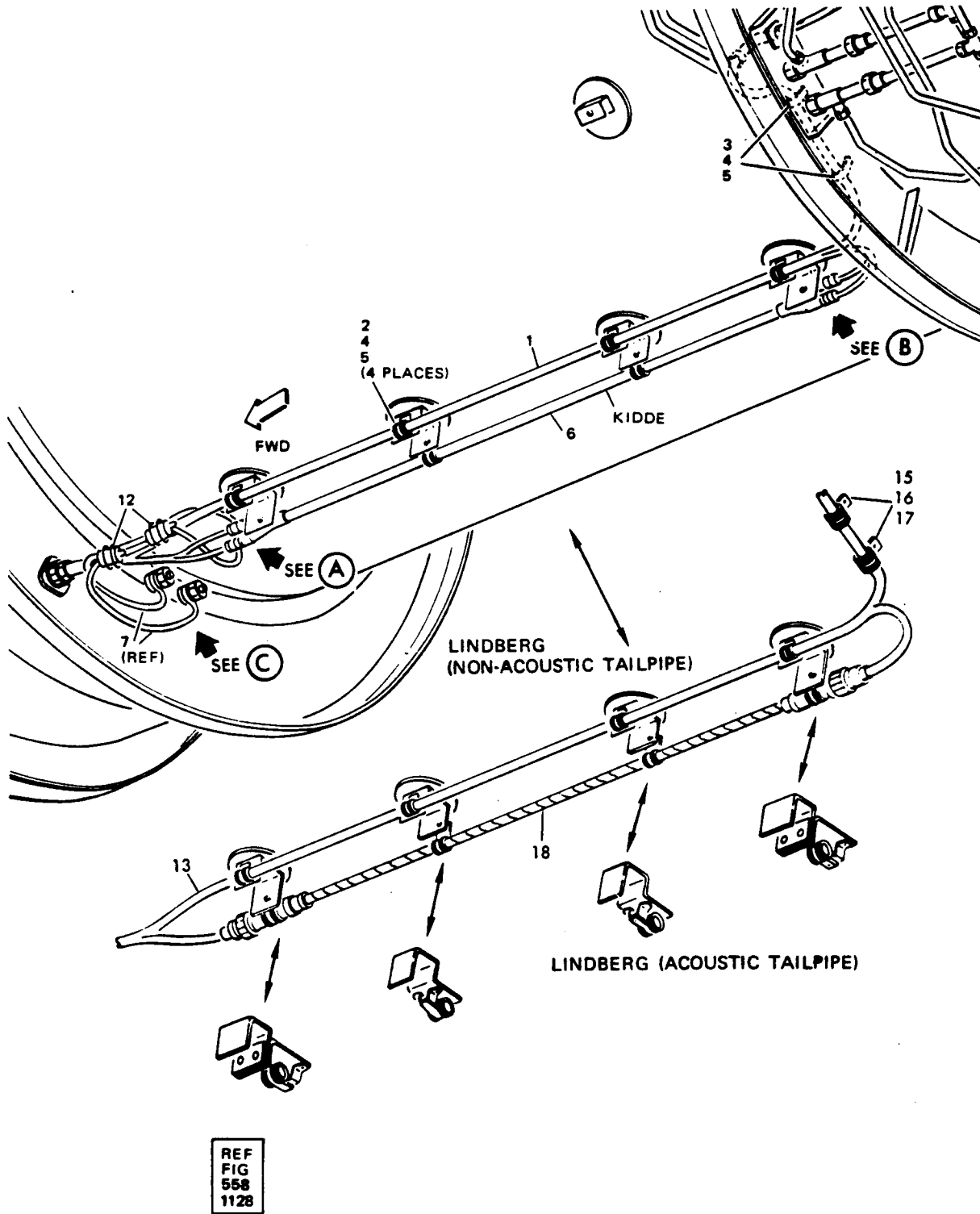
- *[1] TUBE ASSY AND ATTACHING PARTS ARE COMPONENTS OF THRUST REVERSER PLUMBING SYSTEM.
- *[2] CONTROL ASSY AND ATTACHING PARTS ARE COMPONENTS OF THRUST REVERSER PUSH-PULL CABLE SYSTEM.
- *[3] USED ON TELEFLEX CONTROL ASSY
- *[4] USED ON CONTROLEX CONTROL ASSY

- C. Install fire detector element and electrical wiring on engine No. 1 tailpipe extension as follows (Fig. 564):

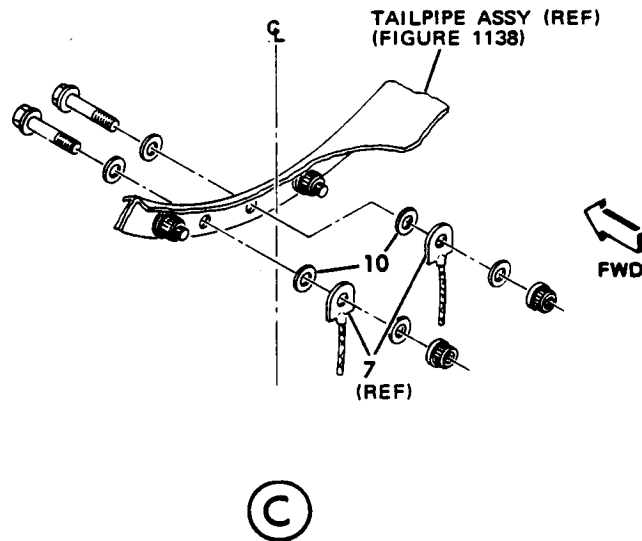
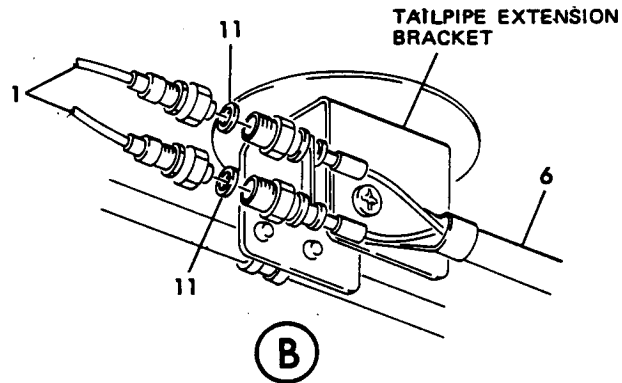
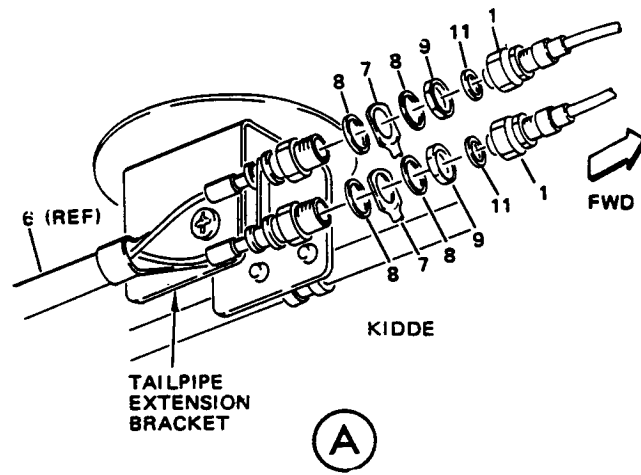
NOTE: This installation is omitted from later airplanes per PRR 32551, and from airplanes modified per SB 26-1017.

Steps (1), (2), (3) and (4) apply to installations with Kidde fire detector systems; steps (5), (6) and (7) apply to installations with Lindberg fire detector systems.

- (1) Install wire bundle assembly (1) with clamps (2 and 3), screws (4), and nuts (5).
 - (2) Install fire detector element (6) in clamps (Fig. 558).
 - (3) Install grounding straps (7) on fire detector connectors, using washers (8) each side of grounding strap terminals, and nuts (9). Lockwire nuts (9). Attach other ends of grounding straps to engine flange as shown, using washers (10) in conjunction with flange attaching parts (Fig. 558). Check that maximum resistance between faying surfaces is 0.0025 ohm.
 - (4) Connect wire bundle assembly to fire detector element and to both engine and thrust reverser wire bundle receptacles using copper gaskets (11) in the following manner:
 - (a) Remove protective cap from electrical connector and check that both connectors are clean and dry.
 - (b) Fit new copper gasket (11) into wire bundle connector.
 - (c) Carefully engage contact pin of bundle connector with element connector receptacle. Tighten bundle connector 50 to 70 pound-inches and lockwire to opposite connector nut.
- CAUTION: TO AVOID TWISTING ELEMENT, HOLD ELEMENT CONNECTOR SECURE WHILE TIGHTENING BUNDLE CONNECTOR.
- (d) Secure grounding straps (7) to wire bundle (1) with spacers (12), using cord suitable for 200°F.
 - (5) Install wire bundle (13) with clamps (14 and 15), screw (16), and nuts (17).
 - (6) Clamp fire detector element (18) in position (Fig. 558).
 - (7) Connect wire bundle assembly to fire detector element and to both engine and thrust reverser wire bundle receptacles.



Fire Detector and Wire Bundle Installation - Engine No. 1
Figure 564 (Sheet 1)



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY	
6.C. (1)	1	WIRE BUNDLE ASSY	61-30406-001		1	
	2	CLAMP *[1]	BAC10HS()		4	
	3	CLAMP *[1]	BAC10HS()		2	
	4	SCREW	BACS12CB3		6	
	5	NUT	BACN10JC3L		2	
	(2)	6	FIRE DETECTOR ELEMENT	10-61096-19		1
	(3)	7	GROUNDING STRAP	BACJ40AD67-12		2
		8	WASHER	AN960C616L		4
		9	NUT	207865		2
		10	WASHER	AN960C516L		2
	(4)	11	GASKET (KIDDE)	209592		4
		12	SPACER	63-9723-3		2
	(5)	13	WIRE BUNDLE ASSY	61-30404-002		1
		14	CLAMP *[1]	BAC10HS()		4
		15	CLAMP *[1]	BAC10HS()		2
		16	SCREW	BACS12CB3-7		6
		17	NUT	BACN10JC3C		2
	(6)	18	FIRE DETECTOR ELEMENT	10-61096-9		1

*[1] SIZE DETERMINED ON INSTALLATION

D. Install potable water pressure plumbing on engine No. 1 as follows:
(See figure 565, view 1.)

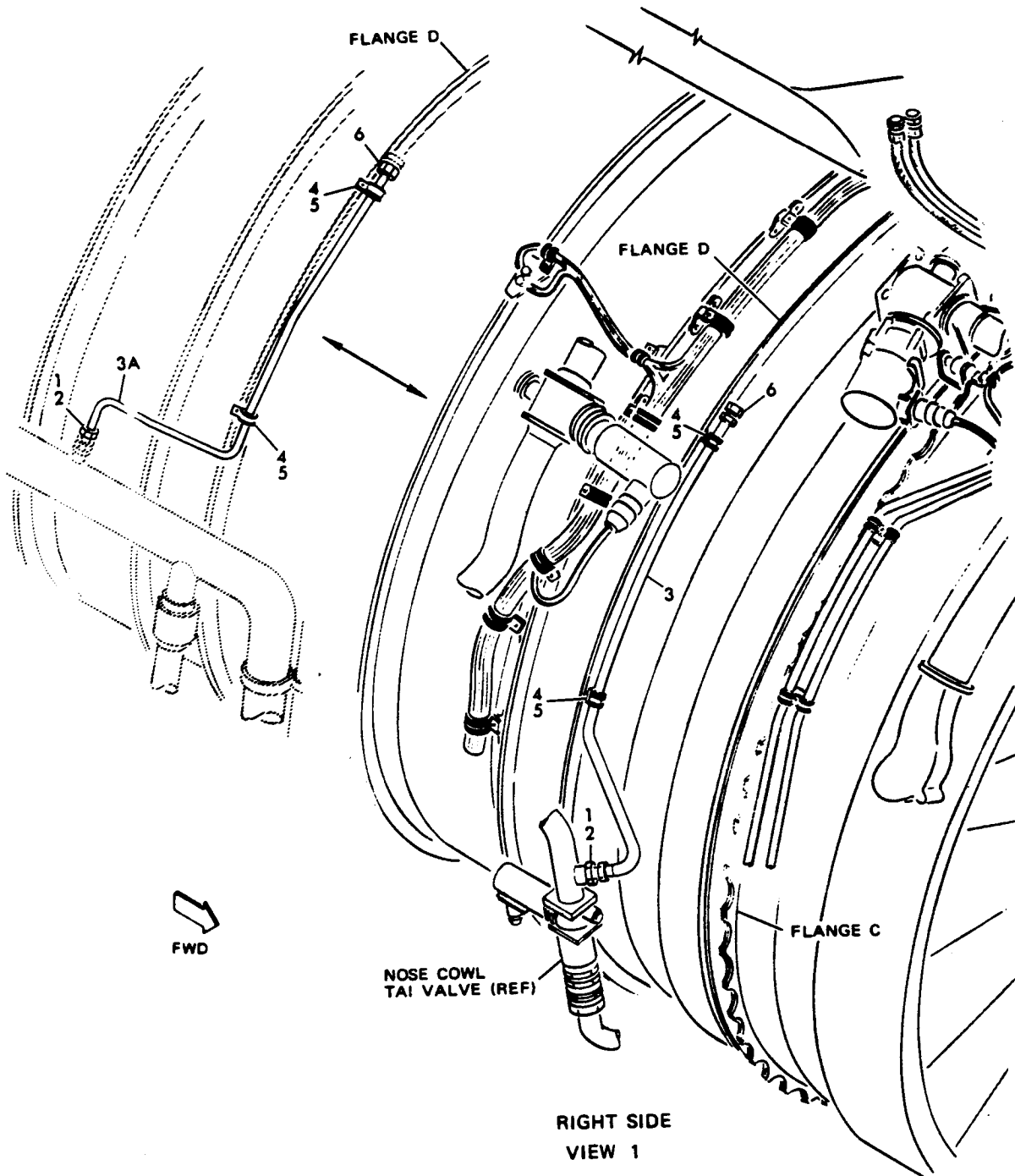
- (1) Install special fitting (1) and gasket (2) in port on TAI duct just above thermostatic valve, or on TAI duct as shown in alternate detail for engines with vortex dissipator gravel protection provisions.
- (2) Locate tube assembly (3 or 3A) and attach to brackets on flange D with clamps (4) and screws (5). Connect hose to fitting (1).
- (3) Install union (6) in upper end of tube assembly.

NOTE: For engines with alternate configuration, install components nose cowl thermal anti-icing system by performing the following steps:

- (4) Locate preassembled filter and valve assembly (7) on right side of flange D and attach to existing brackets with screws (8) and nuts (9).
- (5) Install union (10) with gasket (11) in port on TAI duct just above thermostatic valve, or on TAI duct for engines with vortex dissipator gravel protection provisions. Attach free end of hose assembly of filter and valve assembly (7) to union (10).
- (6) Install elbow (12) in forward port of filter and valve assembly (7).
- (7) Connect TAI hose at top of filter and valve assembly (7).
- (8) Install valve (13) and gasket (14) in duct on left side of engine, just forward of flange F.
- (9) Install elbow (15), washer (16), and nut (17) at valve (13).
- (10) Install bulkhead union (18), washer (19), and nut (20) at bracket on flange D.
- (11) Install hose assembly (21) between elbow (15) and union (18).
- (12) Position tube assembly (22) over engine at flange D. Attach tube assembly at union (18) and elbow (12) and secure to flange D with clamps (23) and screws (24) in three places.

737 POWERPLANT BUILDUP

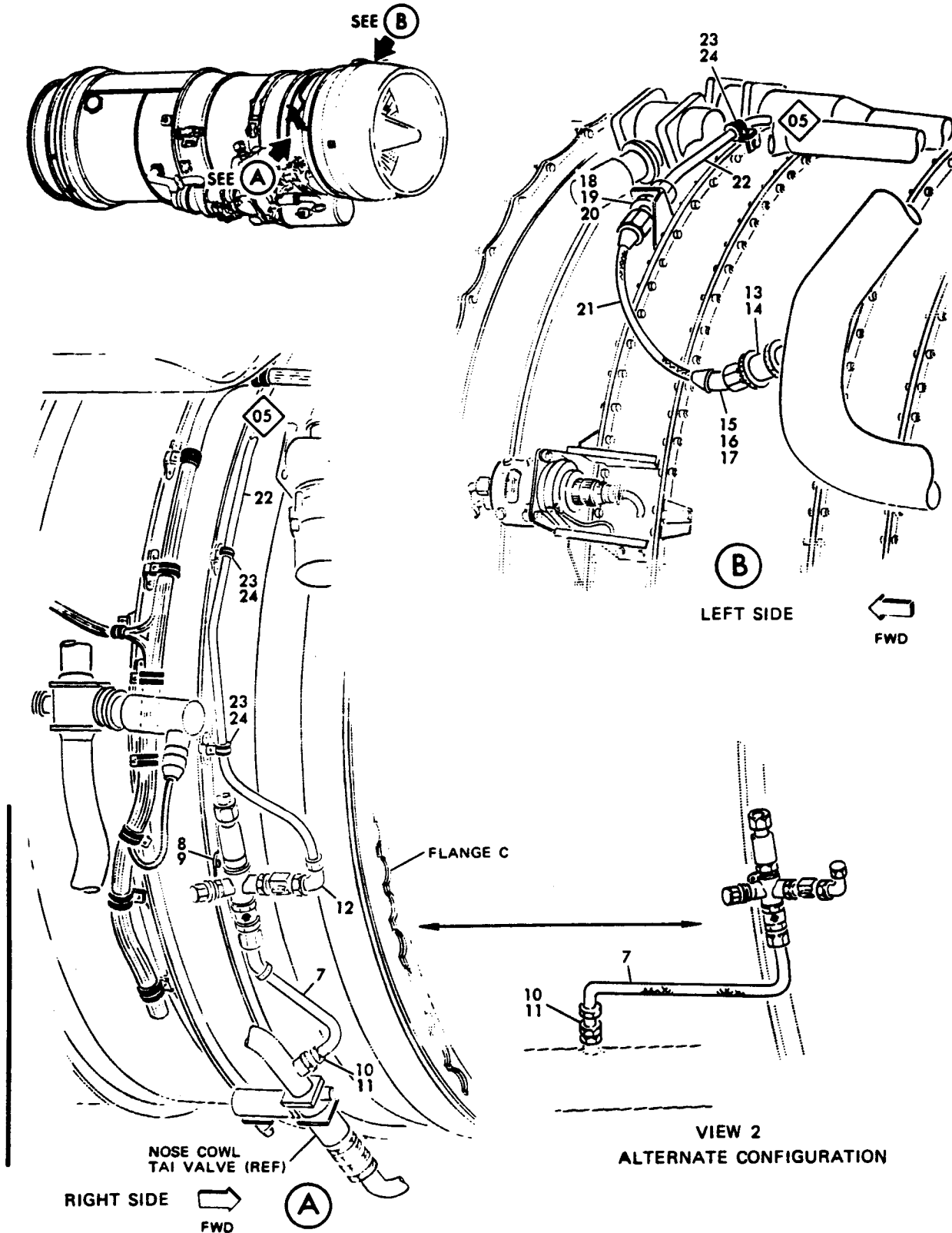
BOEING COMMERCIAL JET OVERHAUL MANUAL



Mar 10/70

Plumbing Installation, Engine Systems
Figure 565 (Sheet 1)

71-03-01
Page 596R



Plumbing Installation, Engine Systems
 Figure 565 (Sheet 2)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
6.D. (1)	1	FITTING, SPECIAL	118-10530		1
	2	GASKET	AN901-8C		1
(2)	3	TUBE ASSEMBLY	65-58135-19 (USED ON 65-58135-18 WITH THRUST REVERSER 65-70950 ONLY)		1
	3	TUBE ASSEMBLY	65-58135-23 (USED ON 65-58135-21 WITH THRUST REVERSER 65-70950 ONLY)		1
	4	CLAMP	BAC10HS06		2
	5	SCREW	BACS12CB3-7		2
(3)	6	UNION	MS21902-6J		1
NOTE: The following parts are used on alternate configuration (nose cowl TAI system).					
(4)	7	FILTER AND VALVE ASSY	(PREASSEMBLED IN PAR. 3.0.)		REF
	8	SCREW	BACS12CB3-8		2
	9	NUT	NAS679A3W		2
(5)	10	UNION	MS21902-8J		1
	11	GASKET	AN901-8C		1
(6)	12	ELBOW	BACE21AW0808J		1
(7)	13	VALVE	8C150		1
	14	GASKET	AN901-10C		1
(8)	15	ELBOW	MS21908-6		1
	16	WASHER	AN960-916		1
	17	NUT	AN924-6		1
(9)	18	UNION	MS21924-8J		1
	19	WASHER	AN960C1216		1
	20	NUT	AN924-8J		1
(10)	21	HOSE ASSEMBLY	16234-1		1
(11)	22	TUBE ASSEMBLY	65-58135-17		1
	23	CLAMP	BAC10HS08		3
	24	SCREW	BACS12CB3-7		3

E. Clamp hydraulic pressure hose on flange "K." Deleted requirement.

Figure 566 Deleted

POWERPLANT BUILDUP

- F. Install striker for thrust reverser flipper door on engine No. 1 as follows: (See figure 566A.)

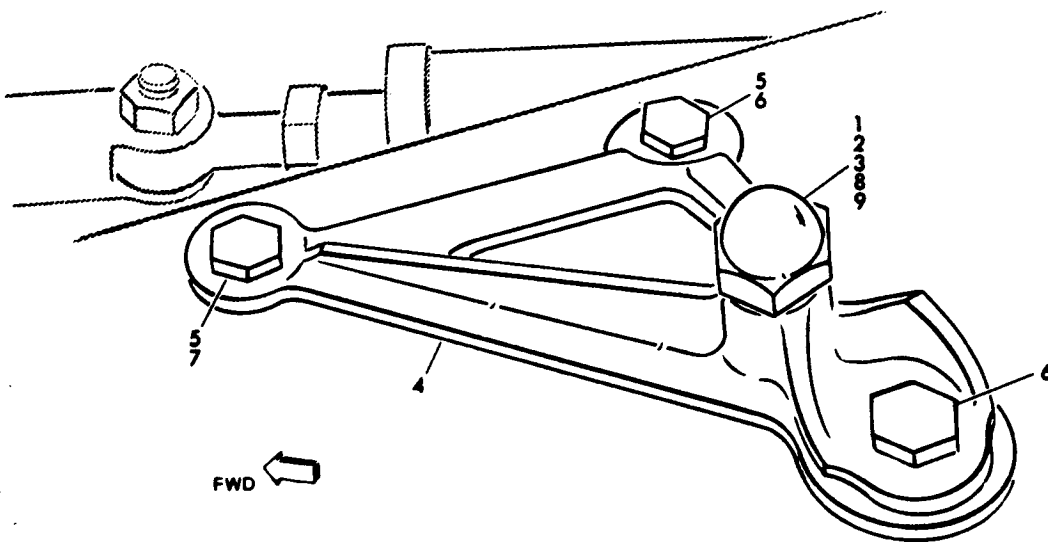
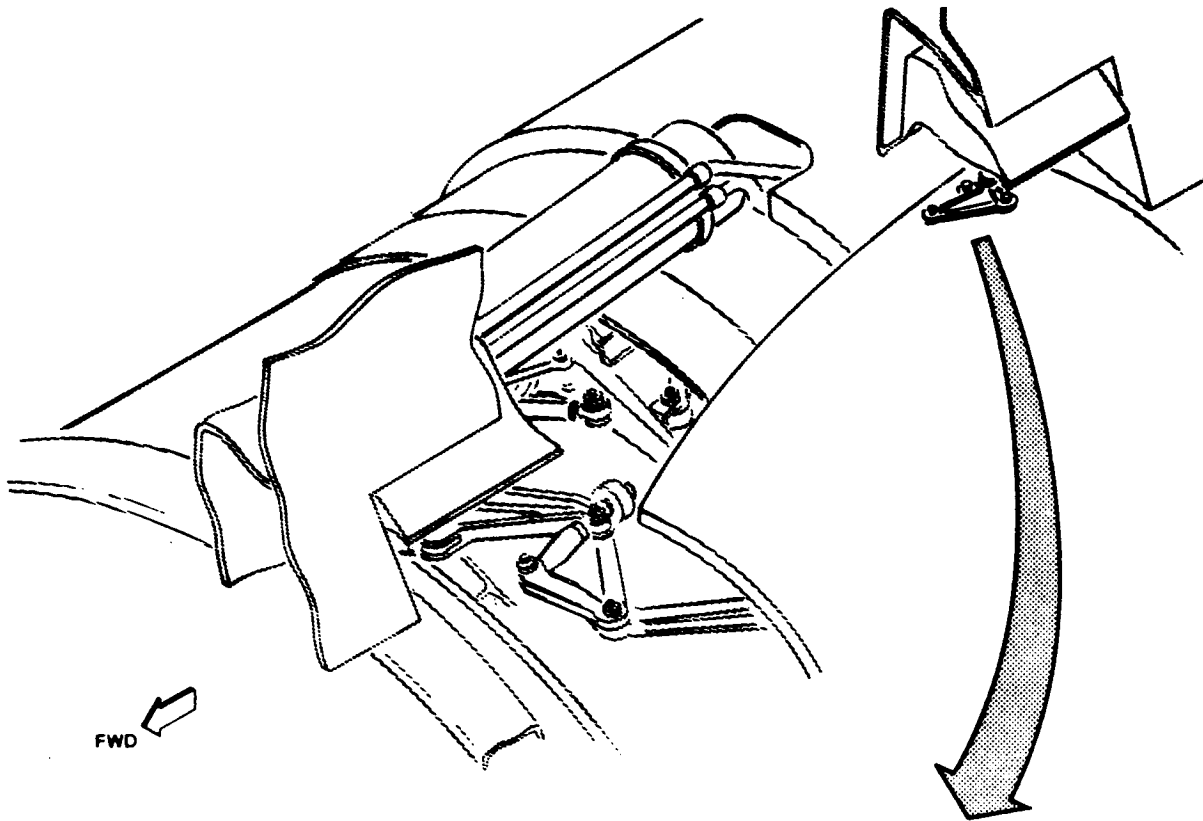
NOTE: This paragraph applies to installations with pneumatically-operated thrust reversers.

- (1) Remove recessed head screw (1). Also remove and retain washer (2) and nut (3) if fitted.
- (2) Install bracket (4) with bolts (5 and 6). If bolt (5) is BACB30FM6A3U, then collar (7), washer (2), and nut (3) are also installed at this stage.

NOTE: Coat threaded surfaces of bolts and nuts prior to installation with EASE-OFF 990 (the Texacone Co.). This note only applies when BACB30FM6A3U bolts are installed.

- (3) Install striker assembly (8) with washers (9).
- (4) Temporarily lockwire striker assembly (8) to bracket (4).

NOTE: For final adjustment of striker assembly, see Maintenance Manual Section 71-00.



REF.
FIG
571 A
1140

Thrust Reverser Flipper Door Striker Installation-Engine No. 1

737


BOEING
COMMERCIAL JET

POWERPLANT BUILDUP

OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
6.F. (1)	1	SCREW	AN509C416R21		Ref
	2	WASHER	BACW10P186C		Ref
	3	BOLT	NAS679C4W		Ref
(2)	4	BRACKET	65-57814-1		1
	4	BRACKET	65-57814-5		1
	5	BOLT	AN3C4A (preferred)		2
	5	BOLT	BACB30FM6A3U		2
	6	BOLT	AN4C14A		1
	7	COLLAR	BACC3026		1
	(3)	8	STRIKER ASSEMBLY	69-49832-1	
	9	WASHER	AN960C516L		4
(4)		No parts required			

Thrust Reverser Flipper Door Striker Installation - Engine No. 1

May 15/68

Figure 566A (Sheet 2)

71-03-01

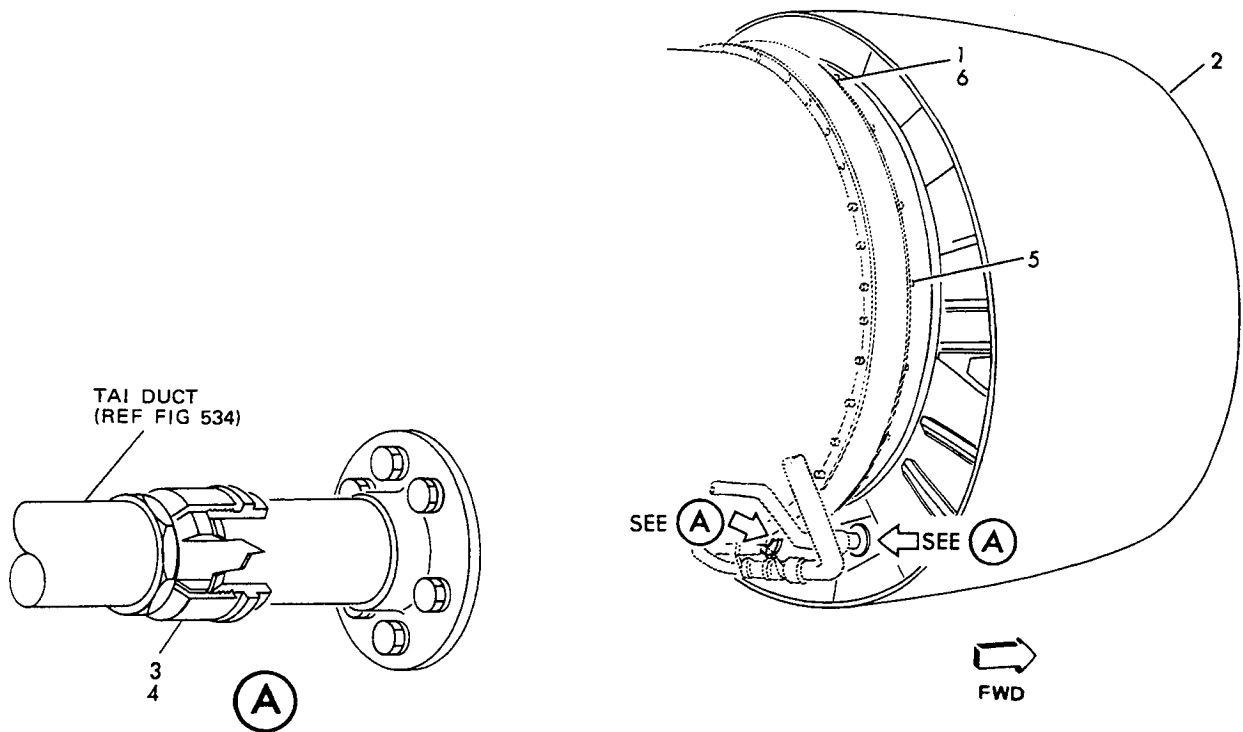
+

Page 598C

7. Final Buildup of Engine No. 2 Only

A. Install nose cowl on engine No. 2 as follows (Fig. 567):

- (1) For 65-60529 series nose cowl only, install 20 bolts (1) loosely in engine flange in positions corresponding to slotted holes in nose cowl attaching ring.
- (2) Install 65-60529 series nose cowl (2) over bolt heads and rotate so bolts butt against narrow end of slots. If 65-85378 series nose cowl is used, position on engine flange and install 23 bolts (1) and 23 washers (6).
- (3) Verify that seal ring (3) in TAI duct and seal ring (4) in vortex dissipator duct are in proper position, and loosely connect ducts to nose cowl (Fig. 534).
- (4) For 65-60529 nose cowl, install three bolts (5) through three unslotted holes in nose cowl attaching ring.
- (5) Tighten bolts (1 and 5) to 125-155 pound-inches.
- (6) Tighten and lockwire TAI fitting and vortex dissipator fitting.



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
7.A. (1)	1	BOLT	MS9148-08 (used with 65-85378 nose cowl) (Fig. 1143)		23
(1)	1	BOLT	MS9148-08 (used with 65-60529 nose cowl) (Fig. 1121)		20
(2)	2	NOSE COWL ASSY	Refer to Fig. 1121 and 1143 for part No.		1
(3)	3	SEAL RING, TAI Duct	NAS595-20E		Ref
	4	SEAL RING, Vortex dissipator duct	NAS595-24E		Ref
(4)	5	BOLT	MS9148-08 (used with 65-60529 nose cowl) (Fig. 1121)		3
(2)	6	WASHER	AN960D516 (used with 65-85378 nose cowl) (Fig. 1143)		23
(5)		No parts required			
(6)		No parts required			

Nose Cowl Installation - Engine No. 2

Figure 567

CAUTION: THRUST REVERSER WEIGHS APPROXIMATELY 386 POUNDS. SUPPORT ENTIRE ASSY DURING INSTALLATION USING SLING F80110.

B. Install thrust reverser (Engine No. 2) (Fig. 568).

CAUTION: BOLT SUBSTITUTION IS NOT PERMITTED UNLESS SPECIFIED.

- (1) Viewing toward the engine tailpipe, position thrust reverser (1) on the engine tailpipe flange with index pin (5A) at location shown in Fig. 567A. Install bolts (2) with heads forward and washers (3) under boltheads. Install nuts, washers (5, 4). Substitution of AN960C516L for item (4) is permitted. One additional washer also may be used.
- (2) Install brackets (6) on forward face of tailpipe flange (Fig. 567A) using bolts, washers, nuts (7, 8, 9). Omit bolts at locations shown.
- (3) Install push-pull control assembly (10) (Teleflex, Fig. 568).

NOTE: Control and attaching parts are components of push-pull cable system.

- (a) Check cable for lubrication. If grease is required, apply Dow Corning 33 (light grade).
- (b) Install forward end of cable of control thru forward fairing and install (11, 12).
- (c) Install aft end of cable of control thru aft fairing and install (13, 14).

CAUTION: USE CARE TO PREVENT CONTAMINATION AND KINKING WHEN INSERTING CABLE THRU SHROUD BULKHEAD.

- (d) Position cable on tailpipe extension using parts (15 thru 20). Do not tighten.
- (e) Loosen nut at B (2 places) and rotate conduit to obtain best possible position and fit. Tighten nut and install lockwire. Tighten all bolts and install lockwire. See Fig. 567B for installation of lockwire at bulkhead.
- (f) Attach push-pull assembly to bellcrank thru access door using parts (21, 22, 23).

- (4) Install push-pull control assembly (10) (Controlex, Fig. 568, 567B).

NOTE: Control and attaching parts are components of the push-pull cable system.

CAUTION: THIS ASSY CONTAINS A FLAT STEEL CENTER CORE WHICH BENDS EASILY IN THE FLAT PLANE BUT RESISTS BENDING IN THE EDGEWISE DIRECTION. DO NOT BEND BY FORCE TO ANY POSITION OR DAMAGE WILL RESULT.

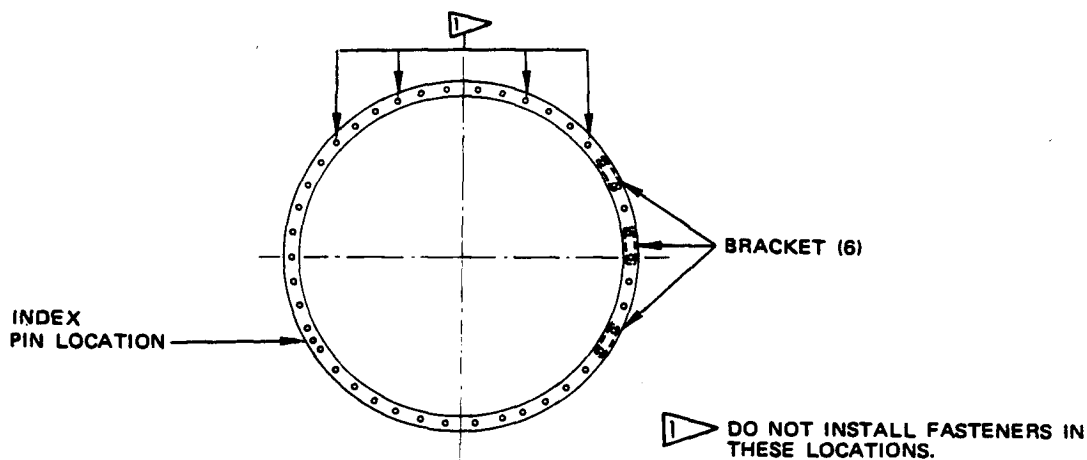
- (a) Install forward end of cable of control (10) thru forward fairing and install (11, 12).
- (b) Attach clamps (15) using (19A, 20). Do not tighten.

CAUTION: CABLE ASSY TELESCOPE ROD MUST BE CENTERED IN SWIVEL WHEN IN UNATTACHED POSITION OR DAMAGE WILL RESULT.

- (c) Insert aft telescopic unit thru aft fairing and install (13, 14). Use four washers at forward side (under head of union) and two washers at aft side (under union nut). Do not tighten bolts (13). Shifting of washers to either side of mounting plate is permitted for adjustment. However, a total of six washers must be used. Tighten union nut.

NOTE: Installation of bracket marked RH is not required for 65-55835-14 and SB 78-1041 configurations.

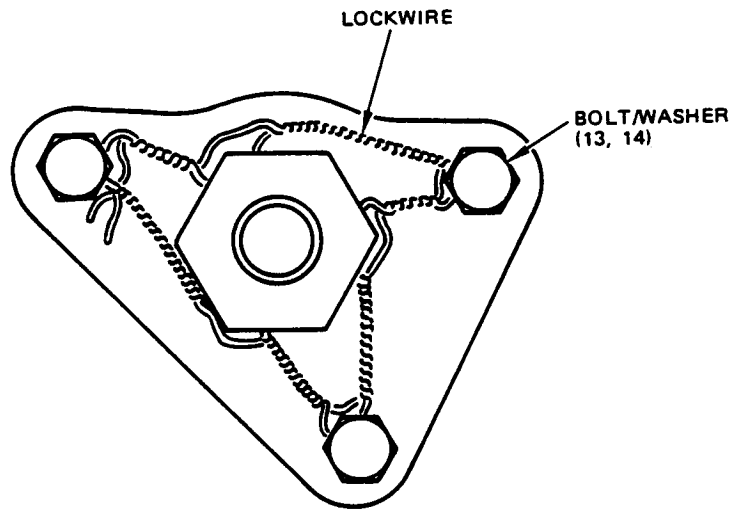
- (d) Attach brackets marked OUT and RH to tailpipe extension (with markings visible) using (19, 19B, 20). Do not tighten.



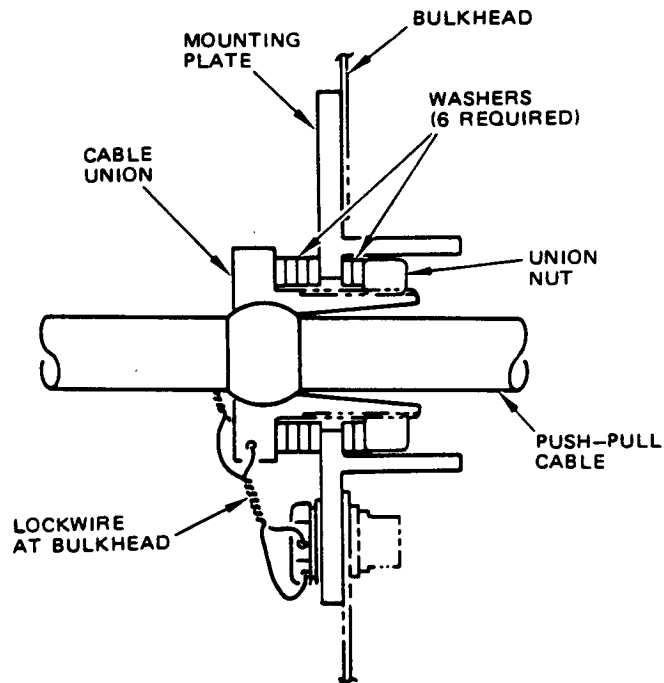
- (e) Provide sufficient loop to cable between connection at aft fairing (Ref step 4a) and bracket marked RH (Ref step 4d) to insure that no preload exists on the telescope rod when it is installed in thrust reverser lever.
 - (f) Adjust cable in brackets to obtain best possible position and fit. Maintain minimum clearance with tailpipe of 0.12 inch by adjusting cable thru clamps. After position is established, tighten all bolts and install lockwire (Fig. 567B).
 - (g) Attach push-pull assembly to bellcrank thru access door using parts (21, 22, 23).
- (5) Attach tubes at bulkhead fittings. Install tubes (24, 25) with clamps (26, 27) and fasteners (28, 29) at three places.

TELEFLEX INSTL
LOCKWIRE END NUT, JAM NUT,
& BOLTS

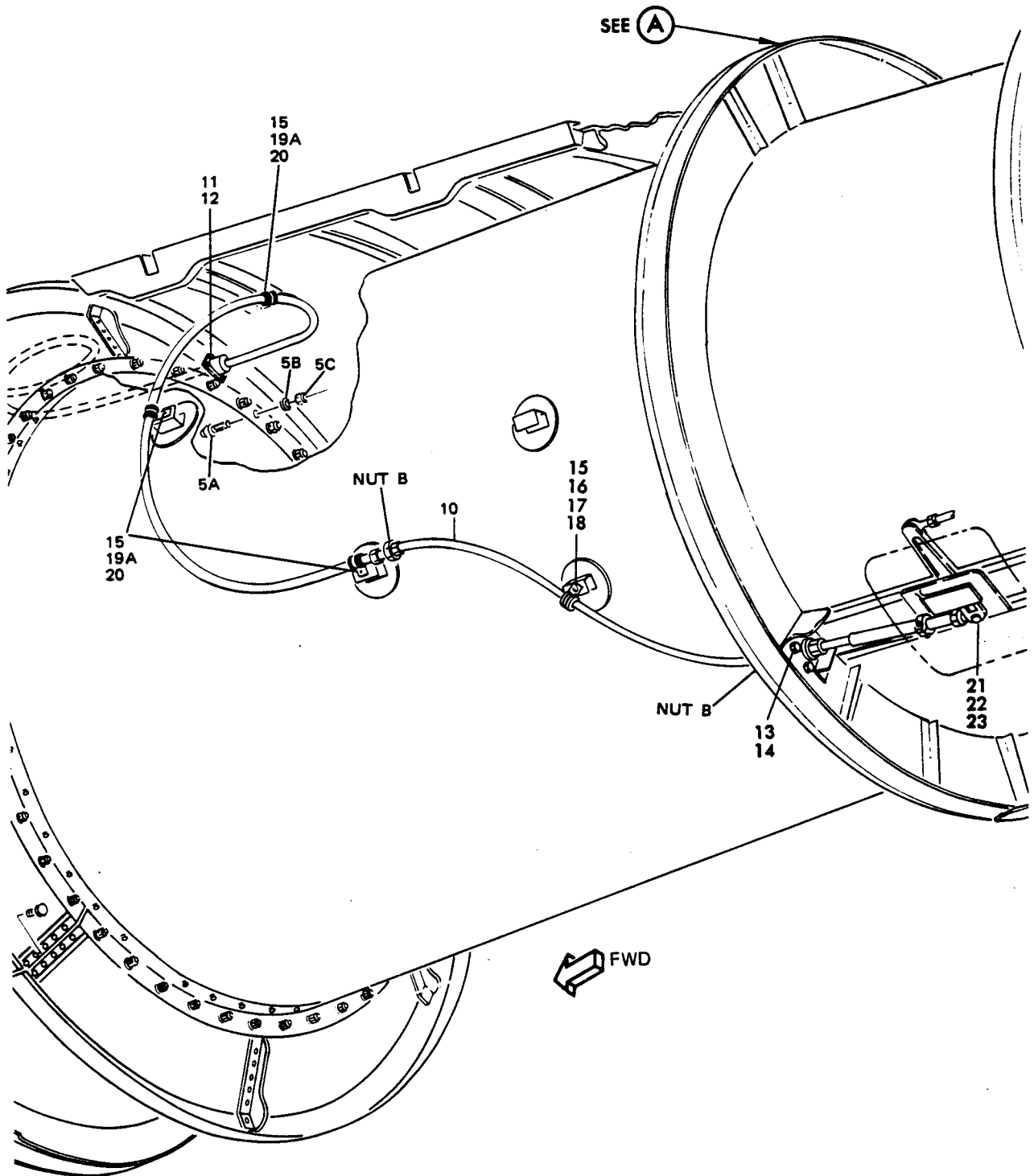
CONTROLEX INSTL
LOCKWIRE UNION & BOLTS



LOCKWIRE DETAILS



SECTION - AFT TELESCOPIC END OF
CONTROLEX CABLE



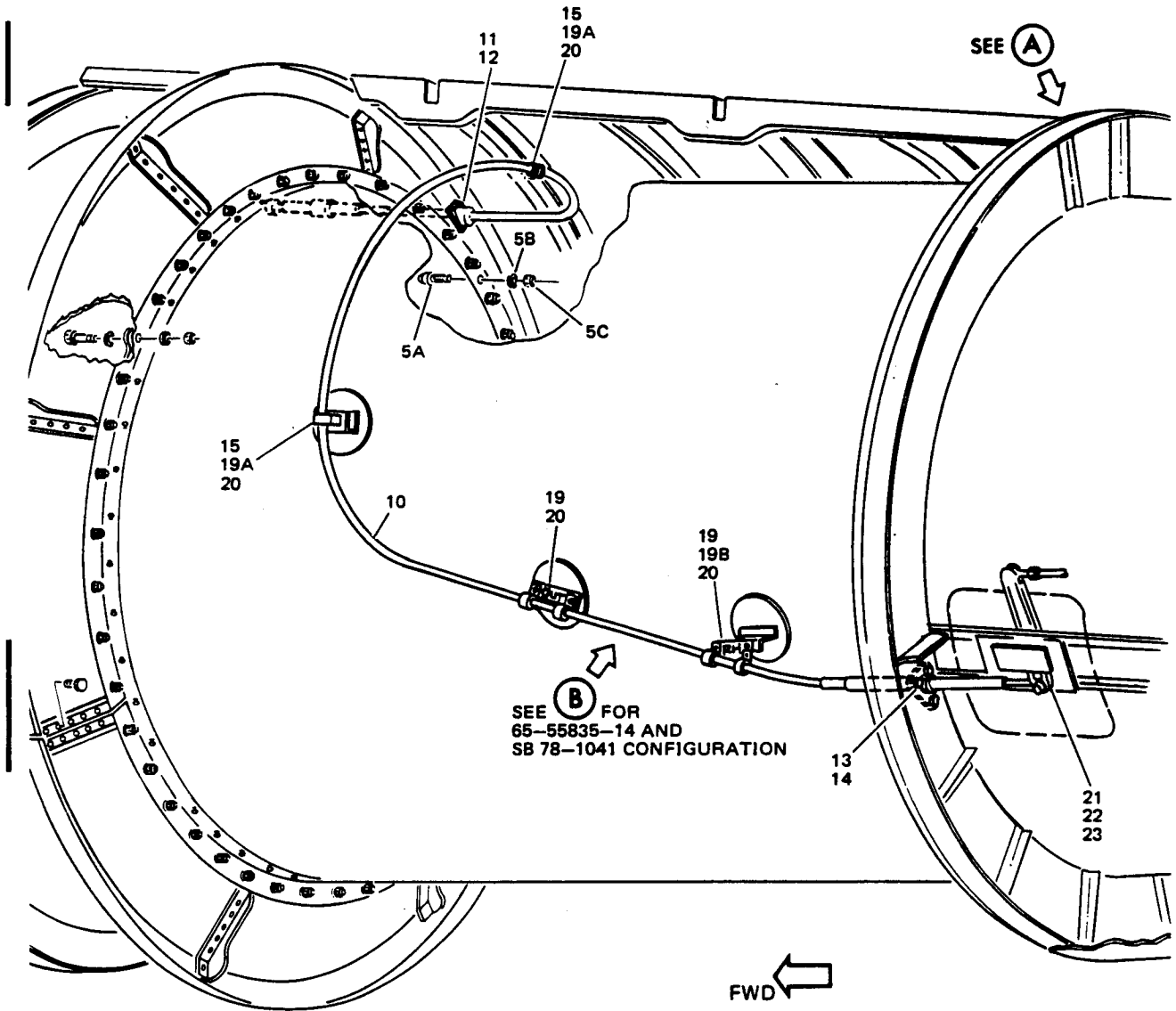
ENGINE NO. 2
(TELEFLEX CONTROL)

REF
FIG
1138

Thrust Reverser, Tailpipe Extension and Shroud Installation
Figure 568 (Sheet 1)

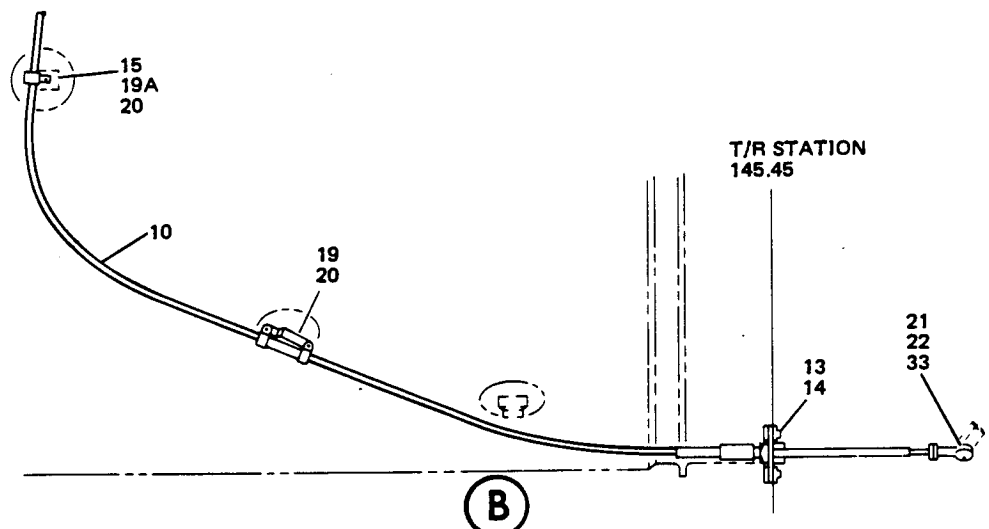
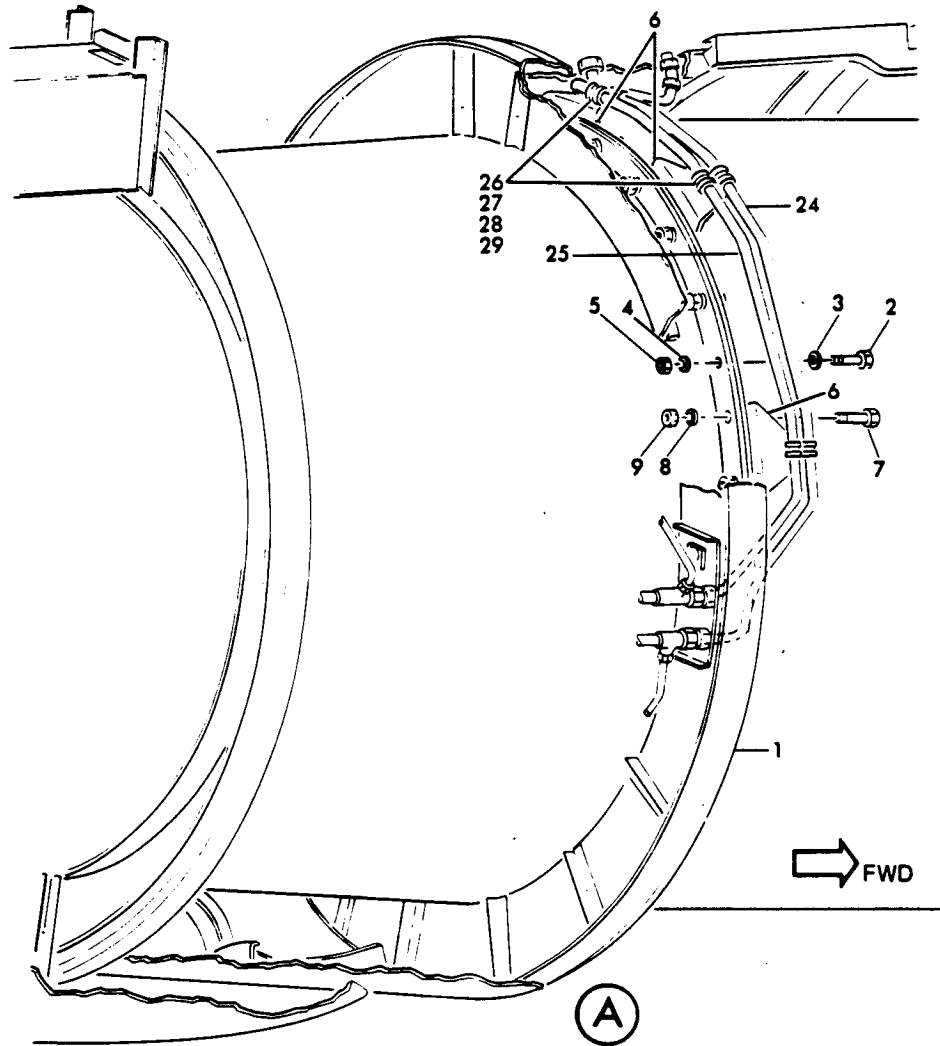
Jun 5/87

71-03-01
Page 598H1



ENGINE NO. 2
(CONTROLEX CONTROL)

REF
FIG
1138



65-55835-14 AND SB 78-1041 CONTROLEX/EANCO PUSH-PULL CABLE CONFIGURATION
ENGINE NO. 2

Thrust Reverser, Tailpipe Extension and Shroud Installation
Figure 568 (Sheet 2)

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY		
7.B. (1)	1	THRUST REVERSER	110-61813-8		1		
	1	THRUST REVERSER	110-61813-28		1		
	1	THRUST REVERSER	110-61813-34		1		
	1	THRUST REVERSER	110-61813-58		1		
	2	BOLT	ACB30LE5U6		32		
	3	WASHER	ACW10BP5ACU		32		
	4	WASHER	N960C516L		32		
	5A	PIN, INDEX	65-70197-4		1		
	5B	WASHER	AN960C106		1		
	5C	NUT	BACN10GW3A		1		
	(2)	6	BRACKET ASSY	69-49854-1		3	
		6	BRACKET ASSY	69-49854-3		3	
		7	BOLT	BACB30LE5U6		6	
		8	WASHER	AN960C516L		6	
		9	NUT	BACN10HR5C		6	
		(3,4)	10	CONTROL ASSY *[2]	25604		1
			10	CONTROL ASSY *[2]	C81662-2		1
			10	CONTROL ASSY *[2]	C81662-3		1
			11	BOLT	BACB30LM4U3		4
	12		WASHER	AN960C416L		4	
	13		BOLT	BACB30LM4HU2		3	
	14		WASHER	AN960C416L		3	
	15		CLAMP *[3]	BACC10HS07		4	
	15		CLAMP *[4]	BACC10HS07		2	
	16		BOLT	BACB30LM3U2		1	
	17		WASHER	AN960-10L		1	
	18		NUT	BACN10JC3C		1	
	19		BOLT	BACB30LM3U3		2	
	19A		BOLT *[3]	BACB30LM3U2		3	
19A	BOLT *[4]		BACB30LM3U2		2		
19B	NUT		BACN10JC3		2		
20	WASHER	AN960C10L		1			
21	BOLT	BACB30LJ4U10		1			
22	WASHER	AN960416L		1			
23	NUT	BACN10JC4C		1			
(5)	24	TUBE ASSY *[1]	65-46858-140		1		
	24	TUBE ASSY *[1]	65-46858-1140		1		
	24	TUBE ASSY	65C21843-4		1		
	25	TUBE ASSY *[1]	65-46858-136		1		
	25	TUBE ASSY *[1]	65-46858-1136		1		
	25	TUBE ASSY	65C21843-2		1		
	26	CLAMP	BACC10HS06		3		
	26	CLAMP	BACC10HS26		3		
	27	CLAMP	BACC10HS08		3		
	27	CLAMP	BACC10HS28		3		
	28	BOLT	BACB30LM4U3		3		
29	NUT	BACN10JC3C		3			

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET

OVERHAUL MANUAL

- *[1] TUBE ASSEMBLIES AND ATTACHING PARTS ARE COMPONENTS OF THRUST REVERSER PLUMBING SYSTEM.
- *[2] CONTROL ASSEMBLY AND ATTACHING PARTS ARE COMPONENTS OF THRUST REVERSER PUSH-PULL CABLE SYSTEM.
- *[3] USED ON TELEFLEX CONTROL ASSY
- *[4] USED ON CONTROLEX CONTROL ASSY

- C. Install fire detector element and electrical wiring, Engine No. 2 tailpipe extension (Fig. 569)

NOTE: This installation is not applicable to airplanes modified per SB 26-1017.

Steps (1), (2), (3), (4) apply to installations with Kidde fire detector systems; steps (5), (6), (7) apply to installations with Lindberg fire detector systems.

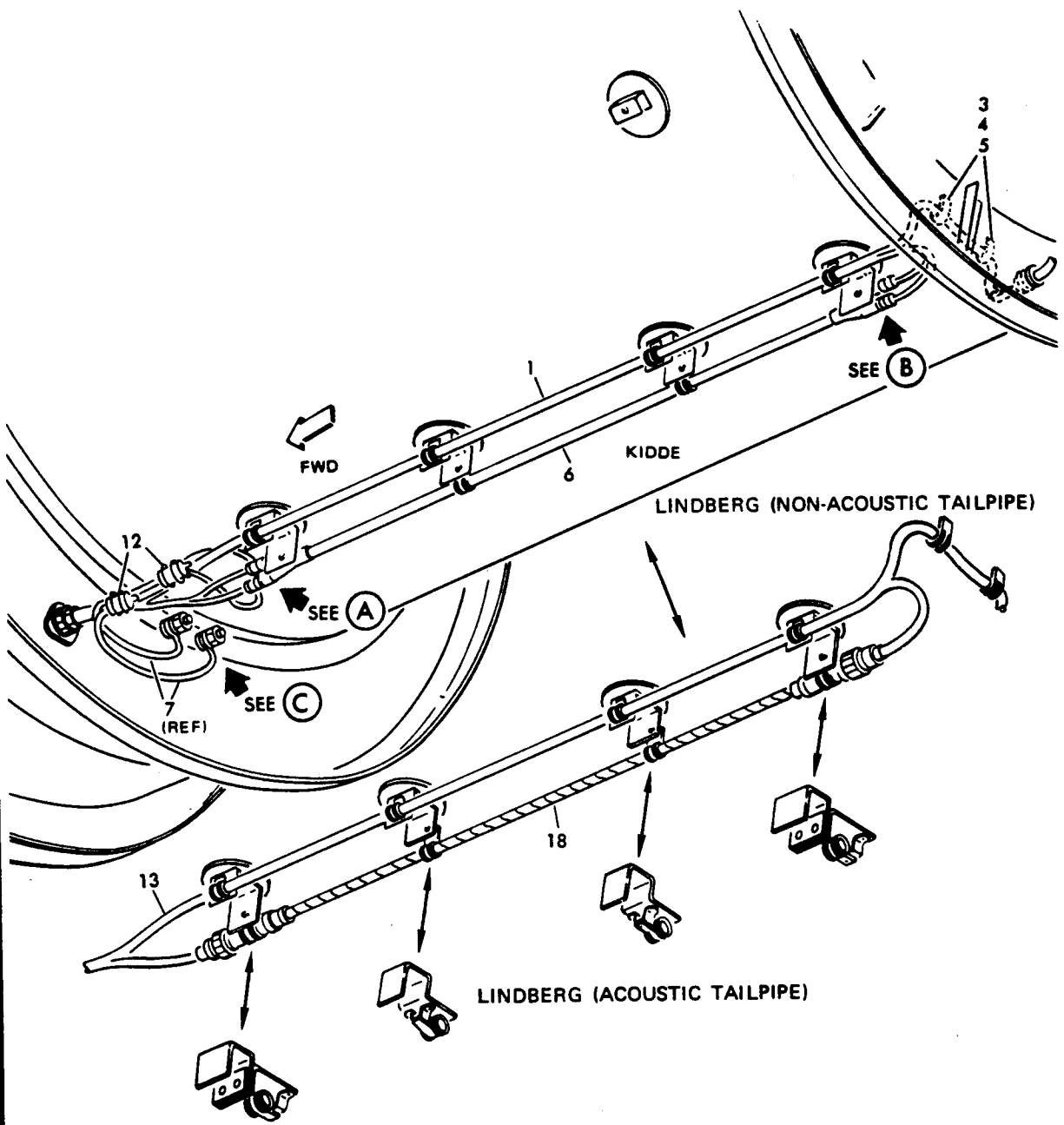
- (1) Install wire bundle (1) with clamps (2, 3), screws (4), and nuts (5).
- (2) Install fire detector element (6) in clamps (Fig. 558).

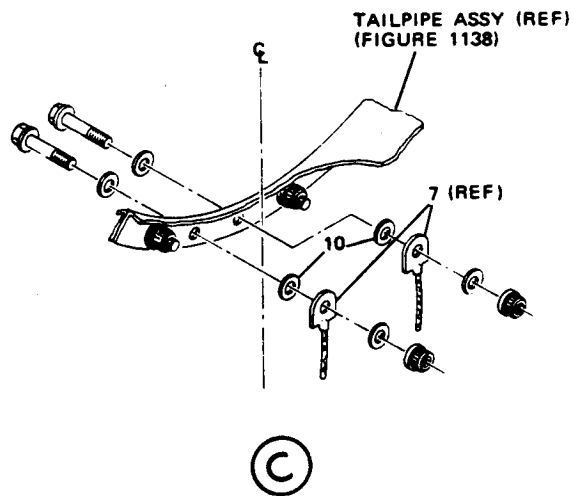
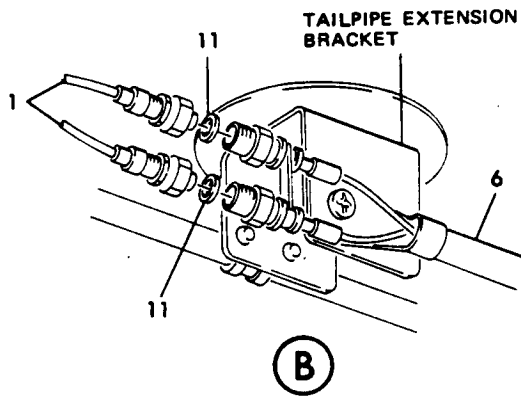
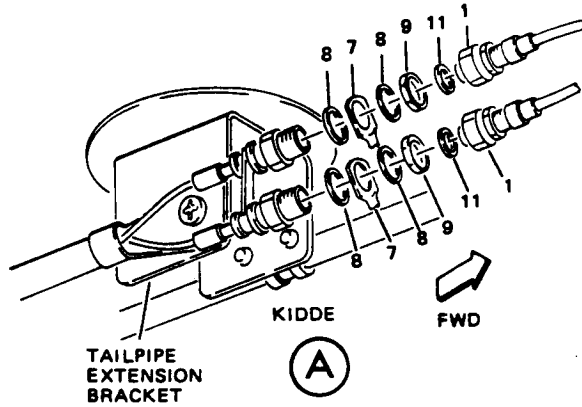
NOTE: Install item (6) with gap in up position.

- (3) Install grounding straps (7) on fire detector connectors using washers (8) each side of grounding strap terminals, and nuts (9). Lockwire nuts (9). Attach other ends of grounding straps to engine flange as shown, using washers (10) in conjunction with flange attaching parts (Fig. 558). Check that maximum resistance between faying surfaces is 0.0025 ohm.
- (4) Connect wire bundle to fire detector element and to both engine and thrust reverser wire bundle receptacles using copper gaskets (11) in the following manner:
 - (a) Remove protective cap from electrical connector and check that both connectors are clean and dry.
 - (b) Fit new copper gasket (11) into wire bundle connector.
 - (c) Carefully engage contact pin of bundle connector with element connector receptacle. Tighten bundle connector 50-70 lb-in. and lockwire to opposite connector nut.

CAUTION: TO AVOID TWISTING ELEMENT, HOLD ELEMENT CONNECTOR SECURE WHILE TIGHTENING BUNDLE CONNECTOR.

- (d) Secure grounding straps (7) to wire bundle (1) with spacers (12), using cord suitable for 200°F.
- (5) Install wire bundle (13) with clamps (14, 15), screw (16), and nuts (17).
- (6) Clamp fire detector element (18) in position (Fig. 558).
- (7) Connect wire bundle assembly to fire detector element and to both engine and thrust reverser wire bundle receptacles.





Fire Detector and Wire Bundle Installation - Engine No. 2
 Figure 569 (Sheet 2)

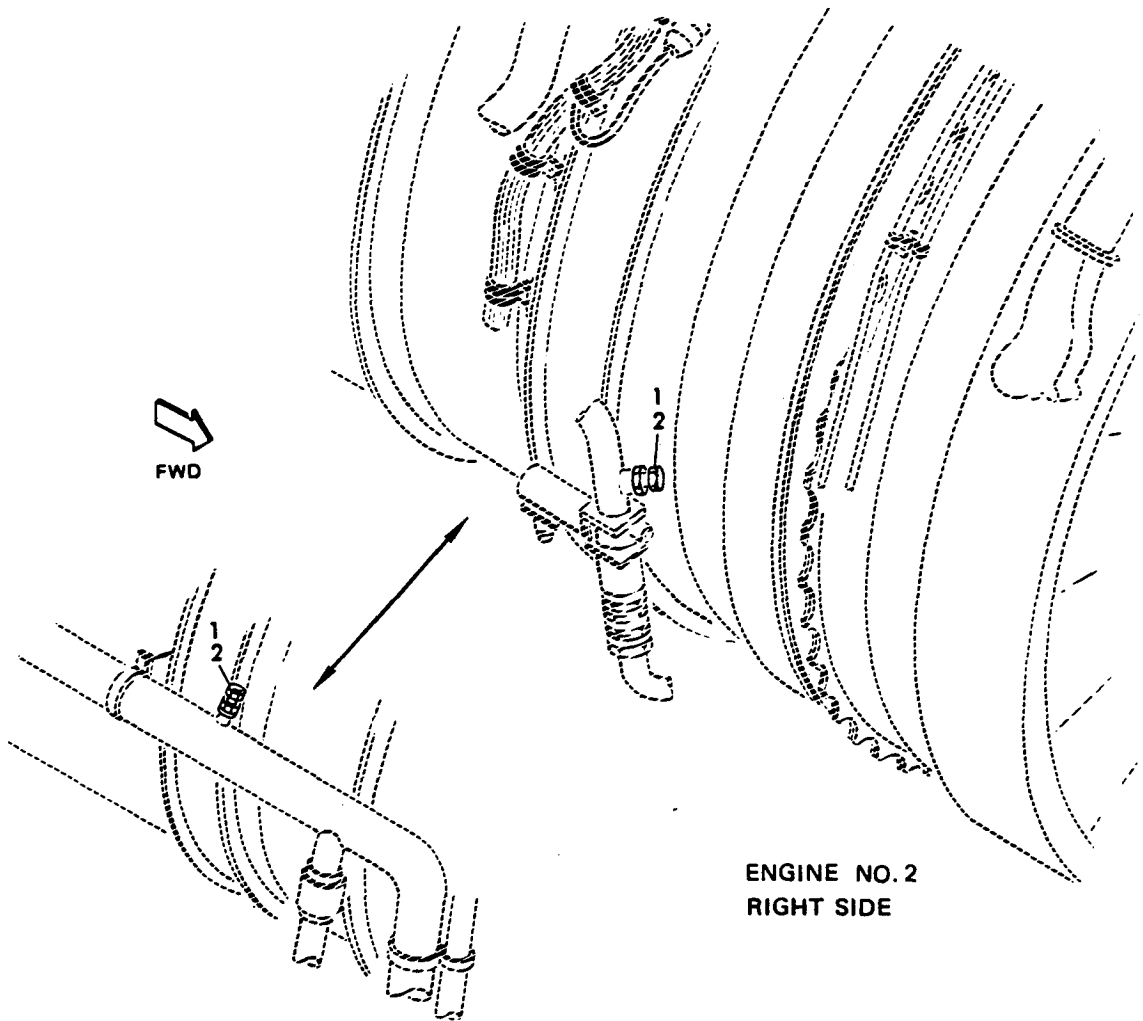
STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
7.C. (1)	1	WIRE BUNDLE ASSY	61-30406-001		1
	2	CLAMP *[1]	BACC10HS ()		4
	3	CLAMP *[1]	BACC10HS ()		2
	4	SCREW	BACS12CB3		6
	5	NUT	BACN10JC3C		2
(2)	6	FIRE DETECTOR ELEMENT	10-61096-19		1
(3)	7	GROUNDING STRAP	BACJ40AD67-12		2
	8	WASHER	AN960C616L		4
	9	NUT	207865		2
	10	WASHER	AN960C516L		2
(4)	11	GASKET (KIDDE)	209592		4
	12	SPACER	63-9723-3		2
(5)	13	WIRE BUNDLE ASSY	61-30404-006		1
	14	CLAMP *[1]	BACC10HS ()		4
	15	CLAMP *[1]	BACC10HS ()		2
	16	SCREW	BACS12CB3-7		6
	17	NUT	BACN10JC3C		2
(6)	18	FIRE DETECTOR ELEMENTS	10-61096-9		1

*[1] SIZE DETERMINED ON INSTALLATION

D. Install plug in TAI duct as follows: (See figure 570.)

- (1) Install plug (1) with gasket (2) in port of TAI duct just above nose cowl TAI thermostatic valve at flange "D." On engines with vortex dissipator gravel protection provisions, install plug (1) and gasket (2) on TAI duct upstream from thermostatic valve.

E. Clamp hydraulic pressure hose on flange "K." Deleted requirements.



STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
7.D. (1)	1	PLUG	BACP20AU8J		1
	2	GASKET	AN901-8C		1

737

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

POWERPLANT BUILDUP

F. Install striker for thrust reverser flipper door on engine No. 2 as follows: (See figure 571A.)

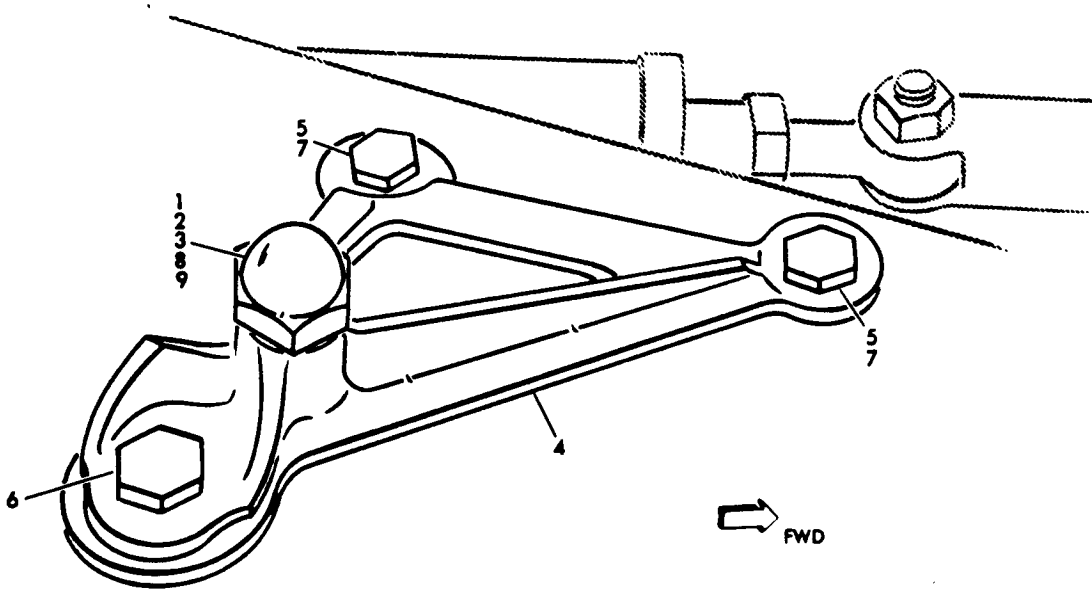
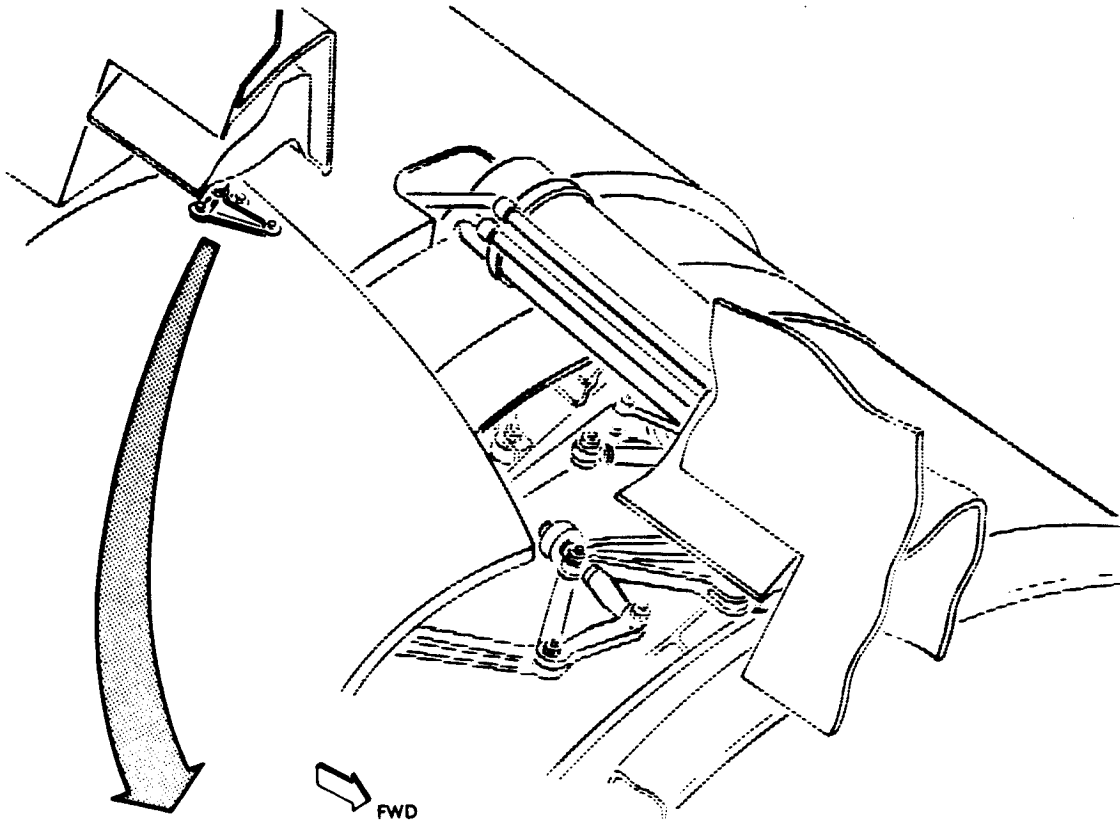
NOTE: This paragraph applies to installations with pneumatically-operated thrust reversers.

- (1) Remove recessed head screw (1). Also remove and retain washer (2) and nuts (3) if fitted.
- (2) Install bracket (4) with bolts (5 and 6). If bolt (5) is BACB30FM6A3U, then collar (7), washer (2), and nut (3) are also installed at this stage.

NOTE: Coat threaded surfaces of bolts and nuts prior to installation with EASE-OFF 990 (the Texacone Co.). This note only applies when BACB30FM6A3U bolts are installed.

- (3) Install striker assembly (8) with washers (9).
- (4) Temporarily lockwire striker assembly (8) to bracket (4).

NOTE: For final adjustment of striker assembly, see Maintenance Manual Section 71-00.



REF.
FIG
566 A
1140

Thrust Reverser Flipper Door Striker Installation - Engine No. 2

737
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

STEP NO.	ITEM NO.	NOMENCLATURE	PART NO.	AIRLINE PART NO.	QTY
7.F. (1)	1	SCREW	AN509C416R21		Ref
	2	WASHER	BACW10P186C		Ref
	3	NUT	NAS679C4W		Ref
(2)	4	BRACKET	65-57814-2		1
	4	BRACKET	65-57814-6		1
	5	BOLT	AN3C4A, preferred		2
	5	BOLT	BACB30FM6A3U		2
	6	BOLT	AN4C14A		1
	7	COLLAR	BACC3026		1
	(3)	8	STRIKER ASSEMBLY	69-49832-1	
9		WASHER	AN960C516L		4
(4)		No parts required			

Thrust Reverser Flipper Door Striker Installation - Engine No. 2

Feb 15/69

Figure 571A (Sheet 2)

71-03-01

Page 5980

+

SYSTEM	APPROVED THREAD COMPOUNDS	
	STRAIGHT THREAD FITTINGS	PIPE THREADS
DEICING OR ANTI-ICING	THREAD COMPOUND MIL-T-5542 OR ANTISEIZE COMPOUND TT-A-580	THREAD COMPOUND MIL-T-5542 OR ANTISEIZE COMPOUND TT-A-580
AIR CONDITIONING	THREAD COMPOUND MIL-T-5542	THREAD COMPOUND MIL-T-5542
FUEL	PETROLATUM VV-P-236 OR ANTISEIZE COMPOUND TT-A-580	ANTISEIZE COMPOUND TT-A-580 OR GREASE MIL-G-6032
LUBRICATING OIL		
HYDRAULIC FLUID	HYDRAULIC FLUID BMS 3-11 OR SKYDROL ASSEMBLY LUBE MCS 352 (MONSANTO)	NO PIPE THREADS USED

OVERHAUL MANUAL

MATERIAL	MIL. SPEC. AND/OR BRAND NAME	USE
Adhesive	Pro-Seal 590M (Coast Pro-Seal and Mfg. Co.) M6249 (United States Rubber Co.)	Attaching nonmetallic parts
Antiseize Compound	Ease-Off 990 (Texacone Inc.) MIL-A-907	Cone bolts (threads only), duct mount bolts, tail plug bolts, and all hot section bolts and threaded fittings
Lubricant	Acryloid HF-825	CSD O-rings
Lubricant	MIL-L-7808	CSD Spline
Lubricant	Esnalube 382 (Elastic Stop Nut Corp. of America)	Apply to nickel alloy bolt threads
Lubricant	Molykote, Type G-N (Dow Corning Co.)	Lube P&W fuel flowmeter fitting
Lubricating Compound	688272-2 (Sundstrand Corp.)	Generator spline, QAD ring, and adapter plate threads
Lubricating Compound	Mix equal volumes of MIL-M-7866 Molybdenum Disulphide and Uni-Temp Grease (Texaco Inc.)	Optional to 688272-2
Lubricating Grease	MIL-G-81322	Tachometer generator shafts
Lubricating Grease	MIL-G-23827	Duct links and engine mounting bolts
Lubricating Grease	MIL-L-4343	Pt2 coupler O-rings
Lubricating Grease	BMS 3-24	Starter spline
Lubricant, Hydraulic	Skydrol Assembly Lube 352 (Monsanto Chemical Co.)	Hydraulic system O-rings

OVERHAUL MANUAL

MATERIAL	MIL. SPEC. AND/OR BRAND NAME	USE
Hydraulic Fluid	BMS 3-11	Service hydraulic pump and lube hydraulic system O-rings
Petrolatum	VV-P-236	Lube straight thread fittings of fuel and oil systems
Engine Oil	P&W 521B	CSD pad cavity

FITS AND CLEARANCES

1. Assembly and rigging fits and clearances are given throughout assembly procedures. For further information refer to Pratt and Whitney JT8D Overhaul Manual.
2. Nearly all threaded fasteners require tightening to a specified torque value. Torque limits and special torque information appear in the following figures:

Figure 601, Torque Limits for Bolts and Nuts
Figure 602, Locking Torque for Self-Locking Nuts
Figure 603, Torque Limits for Flareless Tube Fittings
Figure 604, Torque Limits for Flared Tube Fittings
Figure 605, Torque Limits for Hose End Fittings
Figure 606, Special Torque Information

BOLT OR NUT MAT'L	STEEL, CORROSION RESISTANT STEEL AND TITANIUM *{3}			ALUMINUM (ANODIZED, NOT LUBRICATED) *{3}			
	SELF-LOCKING NUTS, PLATE NUTS AND CASTELLATED NUTS		SELF-LOCKING EXTERNAL WRENCHING NUTS		SELF-LOCKING AND CASTELLATED NUTS		
NUT TYPE	TENSION		SHEAR		TENSION		TENSION
NUT P/N	AN310, AN363 MS20365, BACN10BY, NAS679 AND NAS1021		AN320, MS20364 AND NAS1022		BACN10B		MS20365D AN320D
BOLT PSI	125,000 MINIMUM		125,000 MINIMUM		160,000 MINIMUM		62,000 MINIMUM
TORQUE RANGE (POUND-INCHES)							
THREAD SIZE	*{1}	MAX *{2}	*{1}	MAX *{2}	*{1}	*{1}	MAX *{2}
8-32	12-15	20	7-9	12			
10-32	20-25	40	12-15	25			
1/4-28	50-70	100	30-40	60	65-90	25-35	50
5/16-24	100-140	225	60-85	140	130-180	50-70	100
3/8-24	160-190	390	95-110	240	220-360	85-140	190
7/16-20	450-500	840	270-300	500	370-610	145-240	330
1/2-20	480-690	1100	290-410	660	630-950	230-370	500
9/16-18	800-1000	1600	480-600	960	1000-1300		
5/8-18	1100-1300	2080	660-780	1400	1400-1700		

*{1} These torque ranges apply when tightening nut. When bolt is tightened, use high limit (± 10 percent). Unless specified otherwise, use these torque ranges for listed nuts, plate nuts, bolts and for other bolt-nut combinations of equal or greater tensile strength. See applicable standard drawing for strength and dimensional information.

*{2} Maximum torque allowed for alignment of castellated nut and hole for cotter pin or other safety device insertion.

*{3} Add no lubrication except as required by specific instructions.

* LOCKING TORQUE FOR SELF-LOCKING NUTS					
THREAD SERIES					
FINE			COARSE		
SIZE	TORQUE (POUND-INCHES)		SIZE	TORQUE (POUND-INCHES)	
	**MINIMUM LOCKING	**MAXIMUM LOCKING		**MINIMUM LOCKING	**MAXIMUM LOCKING
4-48	---	3	4-40	---	3
6-40	1.0	6	5-40	1.0	4
8-36	1.5	9	6-32	1.0	6
10-32	2.0	13	8-32	1.5	9
1/4-28	3.5	30	10-24	2.0	13
5/16-24	6.5	60	1/4-20	4.5	30
3/8-24	9.5	80	5/16-18	7.5	60
7/16-20	14.0	100	3/8-16	12.0	80
1/2-20	18.0	150	7/16-14	16.5	100
9/16-18	24.0	200	1/2-13	24.0	150
5/8-18	32.0	300	9/16-12	30.0	200
3/4-16	50.0	400	5/8-11	40.0	300
7/8-14	70.0	600	3/4-10	60.0	400
1-14	92.0	800	7/8-9	82.0	600
1-1/8-12	117.0	900	1-8	110.0	800
1-1/4-12	143.0	1000	1-1/8-8	137.0	900
			1-1/4-8	165.0	1000

* Locking torque is torque required to start nut turning when nut is engaged with at least two full bolt threads protruding beyond locking device of nut and there is no axial load on nut.

** Minimum to maximum locking torque range is used for determining usability or reusability of a self-locking nut and bolt combination. Where there is apparent looseness or excessive torque drag when installing nut on bolt, remove parts and test per this table. Torque values given apply for room temperature.

TUBE			TORQUE RANGE - POUND-INCHES	
SIZE	OD INCH	WALL THICKNESS INCH.	ALUMINUM 6061-T6 CRES MIL-T-8504 MIL-T-8606, MIL-T-8808	CRES MIL-T-6845 (304 1/8 HARD)
-3	3/16	.020 AND OVER	80 - 90	100 - 115
-4	1/4		100 - 115	125 - 145
-5	5/16	.028 AND OVER	135 - 160	170 - 220
-6	3/8		170 - 200	225 - 265
-8	1/2		285 - 340	450 - 525
-10	5/8		370 - 440	540 - 630
-12	3/4		415 - 490	675 - 790

Torque Limits for Flareless Tube Fittings
 Figure 603

Mar 10/70

TUBING OD	FITTING OR NUT SIZE	ALUMINUM ALLOY TUBING, BOLT FITTING OR NUT	STEEL TUBING, BOLT, FITTING OR NUT
INCH		POUND-INCHES	POUND-INCHES
1/8	-2	20 - 30	
3/16	-3		90 - 100
1/4	-4	50 - 65	135 - 150
5/16	-5	65 - 80	180 - 200
3/8	-6	100 - 125	270 - 300
1/2	-8	210 - 250	450 - 500
5/8	-10	300 - 350	650 - 700
3/4	-12	425 - 500	900 - 1000
7/8	-14	500 - 600	1000 - 1100
1	-16	600 - 700	1200 - 1400
1-1/4	-20	680 - 800	1200 - 1400
1-1/2	-24	765 - 900	1500 - 1800
1-3/4	-28	890 - 1050	
2	-32	935 - 1100	

NOTE 1: For AN818 or BAC equivalent coupling nuts, the tubing material governs the torque to be used, regardless of the B-nut material.

NOTE 2: For fitting and boss installations the fittings, nut or bolt material governs the torque to be used. This applies to:

1. Any combination of NAS551 or NAS1236 bolt and NAS552, 553, 554, 555, 1237, 1238, 1239 or 1240 fittings in AND10050 bosses.
2. MS33514 style E and MS33515 fittings in AND10050 bosses.
3. AN924 or equivalent jamnuts on bulkheads.
4. BACS11AB seal installations in AND10050 bosses.

HOSE ASSEMBLIES (TORQUE FOR BOTH STEEL AND ALUMINUM FITTINGS IN POUND-INCHES)		
SIZE	MINIMUM	MAXIMUM
-3	85	100
-4	100	120
-5	155	180
-6	210	250
-8	340	420
-10	440	480
-12	725	850
-16	900	1150
-20	950	1150
-24	1000	1150
-32	1700	2000

INSTL FIG. NO.	INSTALLATION	TORQUE IN POUND-INCHES
	DRAIN - TANK Jiffy drains	270-320
506, 507, 508, 509, 510, 534, 547	TAI, AIR CONDITIONING AND STARTER DUCTS Ball Joint nut: 1) Initial torque 2) Final torque Clamps BACC10DU125C BACC10DU150 BACC10DU150C BACC10DU200X BACC10DU250C BACC10DU300AB BACC10DU300ABE BACC10DP300A BACC10DU300T BACC10DU350T Duct rod end bearing checknut	70-96 380-400 40 40 40 40 45 60 60-65 50 50 50 60-85
521	FORWARD CONE BOLT INSTALLATION Forward cone bolt attach nuts (BACB10BY512 or NAS1022A12)	250-1000 *[1]
522	AC GENERATOR AND CSD INSTALLATION QAD adapter bolts Generator attach nuts Cooling duct clamp - BACC10BN262L-R CSD lug bolt (10-61066-10 or -11 CSD) CSD lug bolt (10-61223-2 CSD)	180-210 160-190 20 90-160 144-180
539	FUEL FLOWMETER INSTL P&W fitting	570-630
546	STARTER AND VALVE INSTL Shutoff valve clamp - BACC10DP300A Guide screws - MS24673-3 Starter scroll attach bolts	35-45 25-30 35-45

Special Torque Information
Figure 606 (Sheet 1)

Mar 1/05

71-03-01
Page 607

INSTR. FIG. NO.	INSTALLATION	TORQUE IN POUND-INCHES
562, 567	NOSE COWL INSTALLATION Attach bolts	125-155
525	WIRE BUNDLE INSTALLATION Wire bundle clamp nuts	25-35
561	NOSE DOME INSTALLATION Attach bolts	100-140
552	CSD PRESSURE SYSTEM PLUMBING OIL-IN fitting to CSD case OIL-OUT fitting to CSD case ON 65-55749-8 ONLY Oil inlet and oil outlet lines to CSD fittings	120-150 90-110 240-360
545	OIL DIFFERENTIAL PRESSURE SWITCH PLUMBING Bolt, Connecting Universal elbow to gearbox	133-147
539A	MAGNETIC CHIP DETECTOR INSTALLATION Plug - B6002 Plug - B6003 Plug - B7670	20-60 30-108 8-40

*[1] Initially tighten nut to a minimum torque of 250 pound-inches. Then check that the cone bolt will rotate when a force of 20 to 25 pounds is applied against its top (threaded) end. If necessary, adjust torque. Do not exceed a torque of 1000 pound-inches.

STUD OR BOLT SIZE	SELF-LOCKING NUT TORQUE (POUND-INCHES)
10-32	20-30
1/4-28	65-70
5/16-24	135-145
3/8-24	180-200

Torque Limits for Electrical Grounds
Figure 607

TESTING

1. Perform test procedures in accordance with trim charts, operating procedures, and adjustment procedures given in the Adjustment/Test sections of the Pratt and Whitney JT8D Engine Overhaul Manual and the applicable Boeing Maintenance Manual.

TROUBLE SHOOTING

1. Refer to Pratt and Whitney Engine Overhaul Manual and the applicable Boeing 737 Maintenance Manual. The Pratt and Whitney Overhaul Manual covers troubles that may be met when testing the basic engine. The Boeing 737 Maintenance Manuals cover troubles that may be met when testing systems that are included in a built-up power plant as it is when installed on the airplane. Use both manuals when testing a built-up power plant.

STORAGE INSTRUCTIONS

1. General

- A. The procedure for preservation, or depreservation (reactivation), of the engine varies, depending upon the length of storage period, the type of preservatives used, and whether or not the compressors may be rotated during the storage period.
- B. Initial steps of long term preservation procedure must be performed immediately following testing. After completing these steps, the engine may be transferred to the storage area for completion of preservation procedure.
- C. Store the engine in a sheltered area where it will not be subjected to extreme temperatures that may cause condensation.

2. Preservation Schedule

CAUTION 1: DO NOT SPRAY OIL OR PRESERVATIVE INTO THE COMPRESSOR OR TURBINE END OF THE ENGINE. THE PRESERVATIVE CAUSES DIRT PARTICLES TO STICK TO THE BLADES AND VANES DURING OPERATION. THIS ALTERS AIRFOIL SHAPE AND AFFECTS EFFICIENCY ADVERSELY.

CAUTION 2: PREVENT EXPOSING DEHYDRATED PACKETS TO THE ATMOSPHERE AS MUCH AS POSSIBLE. SEAL DEHYDRATOR STORAGE CONTAINERS IMMEDIATELY AFTER REMOVING NECESSARY NUMBER OF PACKETS.

CAUTION 3: SILICA GEL DESICCANT IN DEHYDRATING PACKETS IS BLUE WHEN SERVICEABLE AND PINK WHEN SATURATED WITH MOISTURE. USE ONLY SERVICEABLE PACKETS.

A. 0 to 7 Days

- (1) Store engine without preservatives in a dry sheltered area. Store in an area where engine will not be subjected to extreme temperature changes which cause condensation.

B. 7 to 28 Days

- (1) Place 26 pounds of desiccant in engine inlet and tail pipe. Place on racks to prevent contact with engine parts.
- (2) Seal all engine openings. Provide windows in exhaust and inlet covers to allow observation of desiccants.

NOTE: No further preservation is necessary provided relative humidity is maintained at less than 40 percent.

C. 28 to 90 Days

- (1) The engine oil system does not need to be drained or preserved.
- (2) Preserve fuel system, place desiccants in engine cavities, and install plugs in engine openings (Ref Pratt and Whitney JT8D Maintenance Manual).

D. 91 Days or Longer

- (1) Protect engine completely. Preserve engine oil and fuel systems as described in Pratt and Whitney JT8D Maintenance Manual.

3. Preservation of Engines Exposed to External Transport Kit Handling

A. Immediately upon receipt of an engine that has been carried on the outside of an airplane and subjected to high altitude atmospheric conditions, dry the engine as follows:

- (1) Cover the engine entirely with a plastic or canvas bag so arranged as to allow warm, dry air to be pumped into the front of the bag and exhausted out the rear. Hold leakage from the bag (except for its exhaust end) to a minimum. This will allow warm air to pass over and through the engine to remove moisture and ice particles. The air temperature should be 150-200°F, and the flow maintained for 2 to 3 hours.

NOTE: The degree of further preservation will depend on the anticipated duration of engine storage. Refer to previous paragraphs in this subsection for correct procedures.

4. Quick Engine Change (QEC) Kit Storage

A. The QEC kit comprises a number of subkit cartons. They are suitable for storage as received. All the QEC subkit cartons should be stored in a cool, dry place to prevent deterioration of the components.

CAUTION: SHIPPING CARTONS ARE NOT MEANT FOR PROLONGED OUTSIDE STORAGE.

SPECIAL TOOLS, FIXTURES, AND EQUIPMENT

1. To ensure satisfactory installation of all components on engine, the following tools are recommended for use during engine buildup:

NOTE: Equivalent substitutes may be used.

NAME	PART NUMBER	NOTE
Wrench-Pneumatic System Ball Joint Outer Bearing Nut	ST2580-153	Replaced by F72999-1
Wrench-Pneumatic System Ball Joint Retainer	ST2580-163-1 ST2580-163-2 ST2580-163-3	Used for installation/removal of ball joint connectors of the thermal anti-icing and air conditioning ducts.
Wrench-Hydraulic Pump to Engine Attaching Bolt	F72921-1 F72921-2	Superseded by commercial equipment.
Wrench-Starter to Engine Attaching Bolt	SE80-1101	Cancelled
Wrench-Thrust Reverser to Engine Attaching Bolt	SE78-3001	Cancelled
O-Ring Installation Tool	ST848	Used to install O-Rings in hydraulic system components.
Hook Assembly-Engine Controls Flexible Linkage	SE76-1001	Holds engine control cables in stowed position to prevent damage when engine is handled.
Thread Protector-Aft Engine Mount Bolt	F80028-1	Replaced by F70266-14 (Part of F70266-305).
Thread Protector-Engine Mount Bolt	F72733-1	Superseded by F70266-16 (Part of F70266-305).
Plug-Engine Inlet	F72717	Used after installation of nose cowl to prevent foreign matter from entering inlet.
Plug-Engine Exhaust	F72749	Replaced by F80093, -1

NAME	PART NUMBER	NOTE
Sling Assy-Thrust Reverser Instl and Handling	F80110-1	Used during installation/removal of target type thrust reverser.
Dolly-JT8D Engine Transportation and Instl	F80062-55	Used during engine installation/removal and for transporting engine with target type thrust reverser.
Kit - Torque Wrenches	F70198-60	Replaced by commercial equipment
Beam Assy-Engine Handling	F72714-71	Replaced by F72988-2,-3,-4
Attach Beam Set-Engine Handling	F72988-2, -3, -4	F72988-2 or -3 is used with F80219-1, -2 for engine installation/removal. F72988-2 may also be used for shop handling. F72988-4 is used for shop handling only.
Thread Protector-Engine Mount Bolt Set	F70266-305	F70266-14 (Part of -305) used on aft engine mount bolts. Replaces F80028-1. F70266-16 (Part of -305) used on Fwd engine mount bolts. Superseded F72733-1.
Wrench Set-Pneumatic System Ball Joint	F72999-1	Used to install/remove outer bearing nuts in the pneumatic system. Replaces ST2580-153.
Plug-Engine Exhaust	F80093,-1	Replaces F72749
Adapter-CSD and AC Generator	F70333-24, -25	Used to install/remove CSD or AC generator. Replaces F80011-1,-16.

ILLUSTRATED PARTS LIST

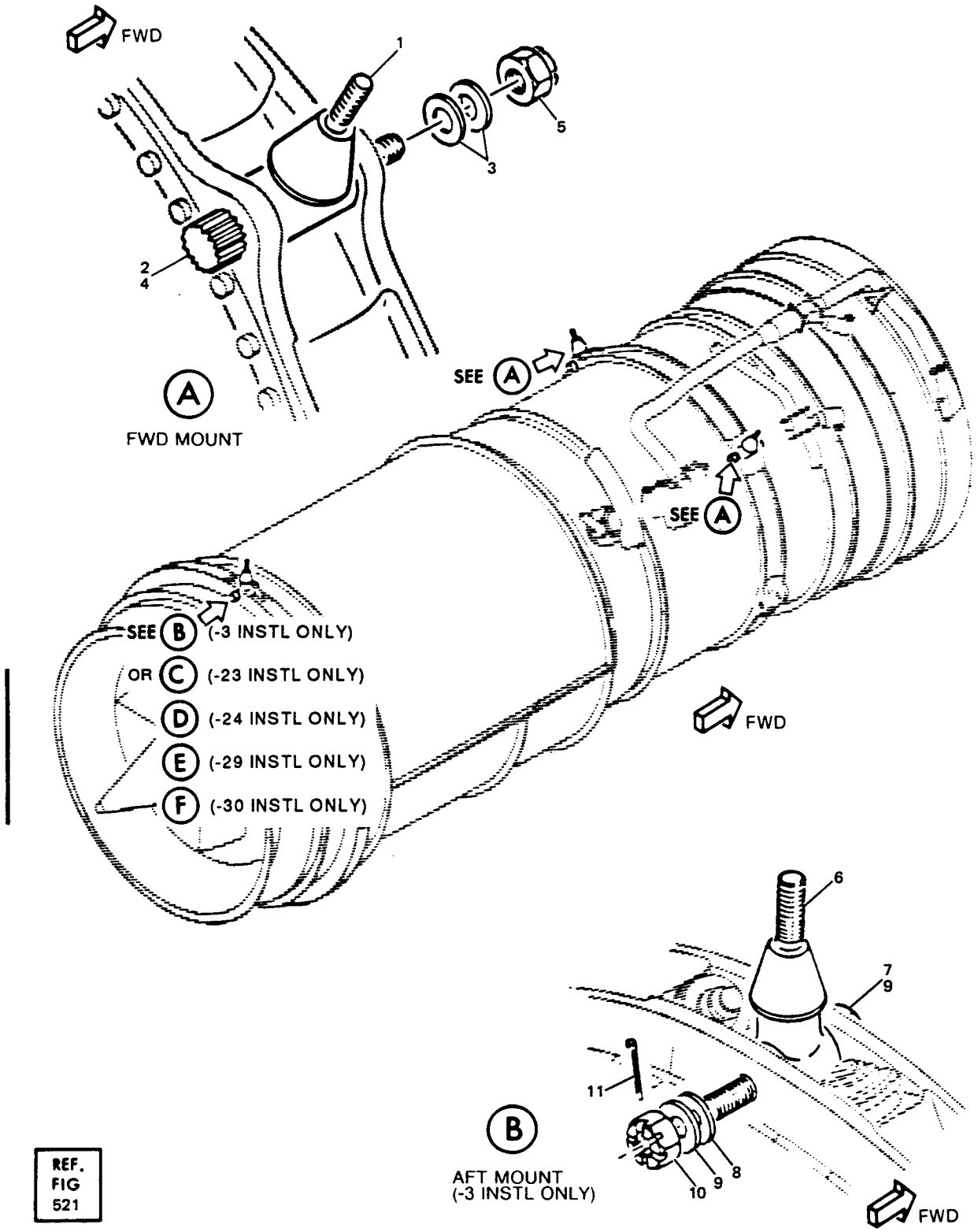
1. General

- A. This section lists all parts supplied in a Quick Engine Change (QEC) kit and used to build a basic engine into a functional power plant.
- B. Deleted.
- C. The part numbers shown reflect the latest information available at the time the revision is prepared. Superseded non-standard parts are not listed. The appropriate engineering drawing should be consulted before substituting any such parts into a power plant.
- D. Standard parts (AN, MS, NAS) listed are those specified on Boeing engineering drawings. These parts may become obsolete or superseded. Refer to BAC Standards Manual No. D.950 for a listing of suitable substitutes for standard parts.
- E. The Illustrated Parts Index lists the installations used on power plant assemblies.

NOTE: Refer to the 1200 page block to determine the installations and configurations applicable to each airline.

INSTALLATION DRAWING NO.	FIG NO.	TITLE
65-19592	1101	ENGINE MOUNT CONE BOLT INSTL
65-19684	1102	ENGINE NOSE DOME INSTL
65-20996	1103	STARTER AND VALVE INSTL
65-21931	1114A	AC GENERATOR AND CONSTANT SPEED DRIVE INSTL
65-21950	1104	N1 AND N2 TACHOMETER GENERATOR INSTL
65-22249	1105	FUEL FLOW METER INSTL
65-22454	1106	ENGINE VIBRATION PICKUP INSTL
65-25736	1107	OIL QUANTITY TANK UNIT INSTL
65-27808	1108	THRUST REVERSER TAIL PLUG INSTL
65-45798	1109	BRACKET INSTL (BOEING)
65-45799	1109	BRACKET INSTL (PRATT AND WHITNEY)
65-46851	1111	AIR CONDITIONING DUCT INSTL
65-46852	1112	CABIN AIR AND STARTER DUCT INSTL
65-46853	1113	ENGINE AIR CONDITIONING SYSTEM PRECOOLER INSTL
65-46858	1138	ENGINE THRUST REVERSER PLUMBING INSTL
65-48361	(Ref)	WIRE BUNDLE AND TUBING INSTL
65-50571	1114	AC GENERATOR AND CONSTANT SPEED DRIVE INSTL
65-50882	1115	ENGINE OIL PRESSURE AND LOW PRESSURE WARNING TRANSMITTER AND SWITCH INSTL
65-51534	1110	ENGINE CONTROLS LINKAGE INSTL
65-52261	1116	ENGINE DRAIN SYSTEM DRAIN FITTING INSTL
65-52971	1117	ELECTRICAL INSTL
65-54476	1118	ENGINE DIFFERENTIAL OIL PRESSURE SWITCH INSTL
65-55475	1119	ENGINE DRAIN SYSTEM PLUMBING INSTL
65-55749	1120	CONSTANT SPEED DRIVE PLUMBING INSTL
65-55750	1121	NOSE COWL INSTL
65-55835	1138	PUSH-PULL CABLE INSTL (REF ONLY)
65-56646	1122	AC GENERATOR COOLING DUCT INSTL
65-58122	1123	CONSTANT SPEED DRIVE FILTER INSTL
65-58125	1124	NOSE COWL THERMAL ANTI-ICING VALVE INSTL
65-58126	1125	AIR CONDITIONING SENSING AND THERMAL ANTI-ICING PLUMBING INSTL
65-58135	1126	ENGINE SYSTEMS PLUMBING INSTL
65-58145	1127	HYDRAULIC SYSTEM PLUMBING INSTL
65-58234	1128	ENGINE FIRE AND OVERHEAT DETECTOR INSTL
65-65919	1108	THRUST REVERSER TAIL PLUG INSTL
65-69659	1142A	METALCAL INSTL
65-70950	1138	THRUST REVERSER, TAILPIPE EXTENSION, AND SHROUD INSTL
65-77451	1125	VORTEX DISSIPATOR VALVE AND PRESSURE SWITCH INSTL
65-57815	1140	THRUST REVERSER FLIPPER DOOR STRIKER INSTL
65-85379	1143	FAT LIP NOSE COWL INSTL

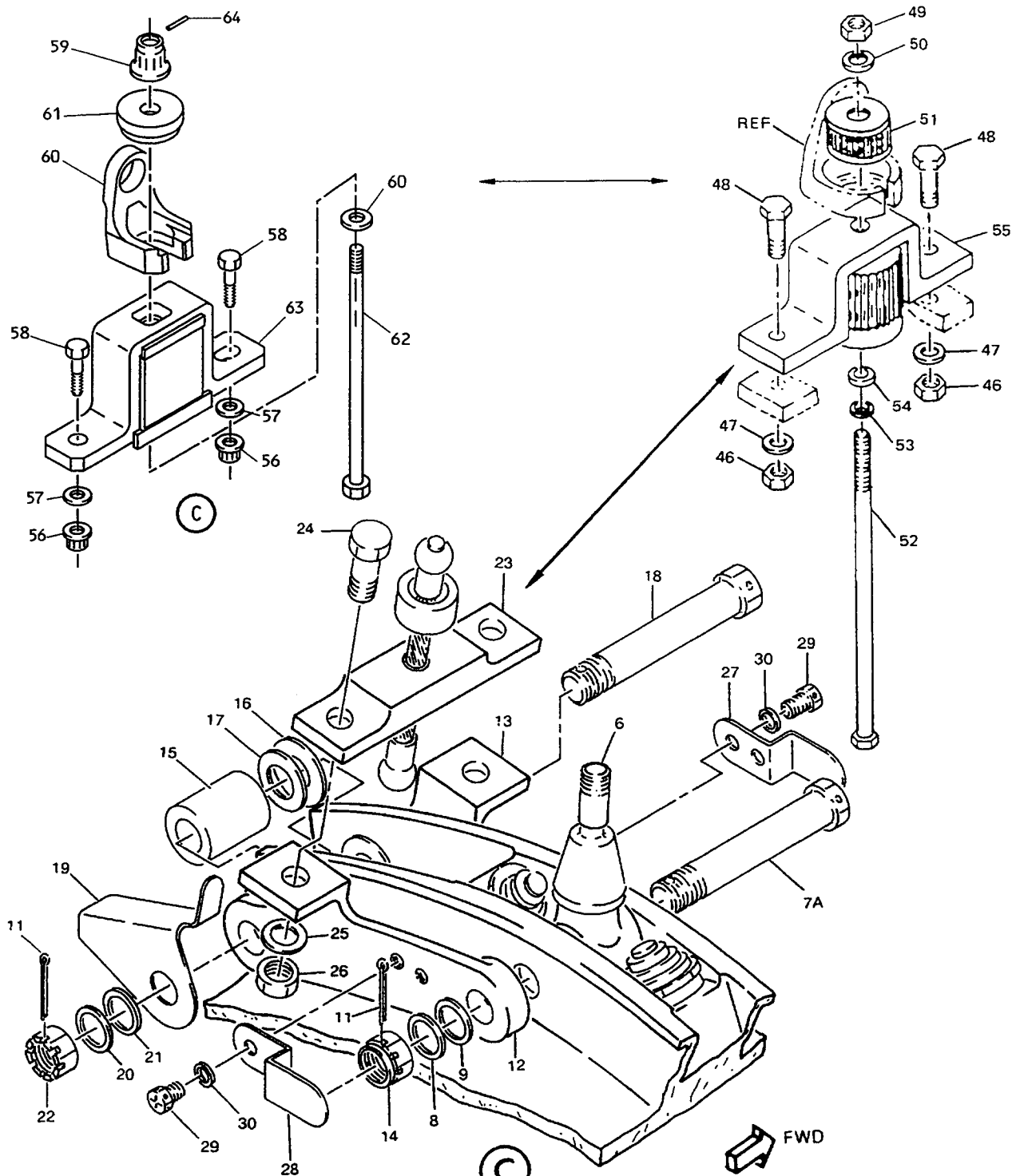
INSTALLATION DRAWING NO.	FIG NO.	TITLE
65C13071	1130	RIPPLE ATTENUATOR INSTL
69-15863	1130	HYDRAULIC PUMP INSTL (ABEX)
69-15863	1130A	HYDRAULIC PUMP INSTL (INTEGRAL HYDRAULIC DAMPER)
69-15935	1131	ENGINE OIL TEMPERATURE TRANSMITTER INSTL
69-21612	1132	OIL TANK PRESSURE FILLING FITTING INSTL
69-28727	1133	HYDRAULIC PUMP INSTL (VICKERS)
69-36670	1134	FUEL HEATER AIR EXHAUST DUCT INSTL
69-37478	1135	CONSTANT SPEED DRIVE OIL COOLER INSTL
69-42802	1111	DIFFERENTIAL PRESSURE REGULATOR INSTL
69-45172	1136	CASE DRAIN HYDRAULIC PUMP FILTER INSTL
69-48936	1137	ENGINE GEARBOX OVERBOARD BREATHER FITTING INSTL
69-50911	1141	LEFT-HAND COWL PANEL HOOK INSTL
69-56334	1139	CHIP DETECTOR INSTL
69-60143	1142	METALCAL INSTL



REF.
FIG
521

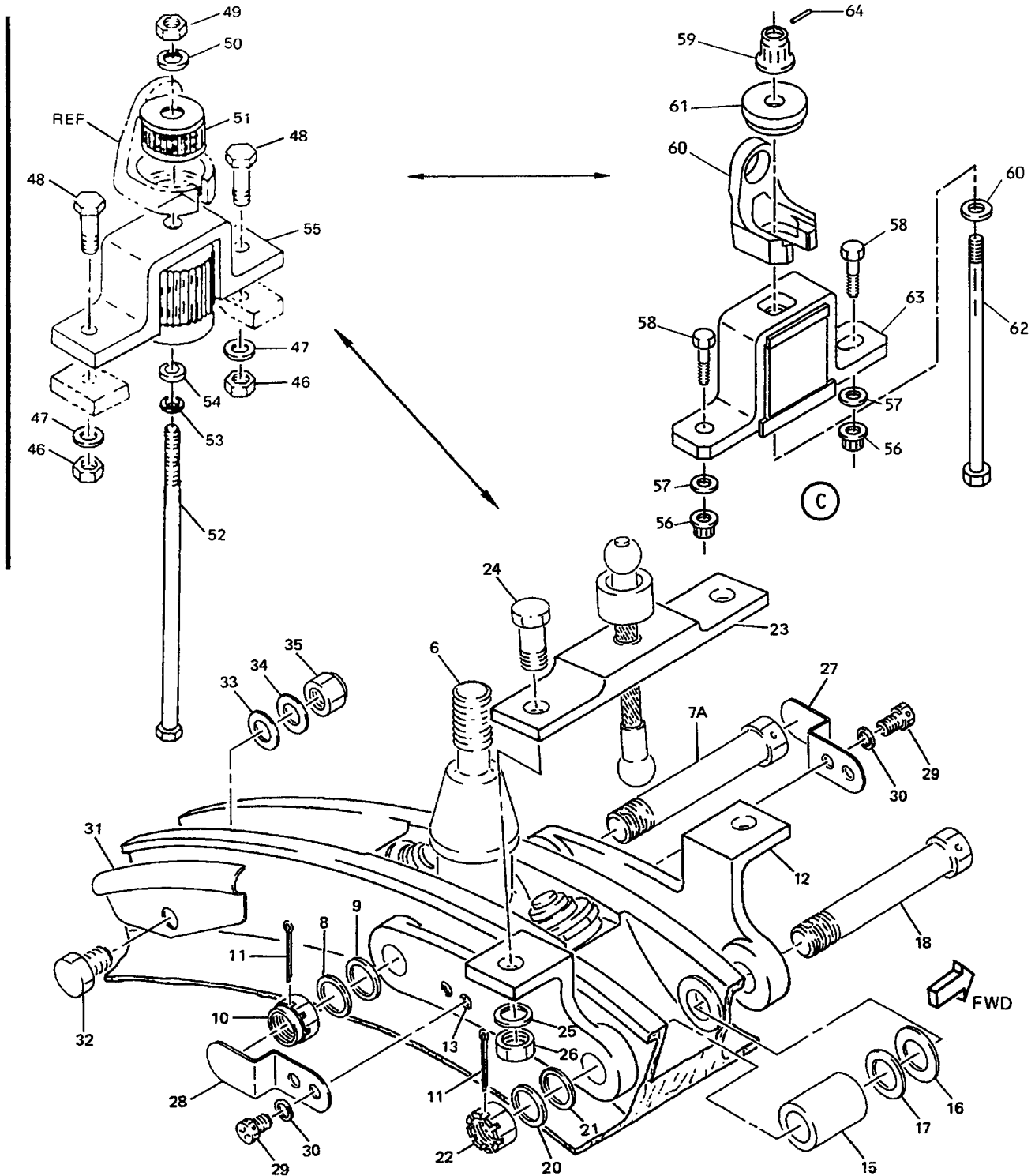
B
AFT MOUNT
(-3 INSTL ONLY)

Engine Mount Cone Bolt Installation
Figure 1101 (Sheet 1)

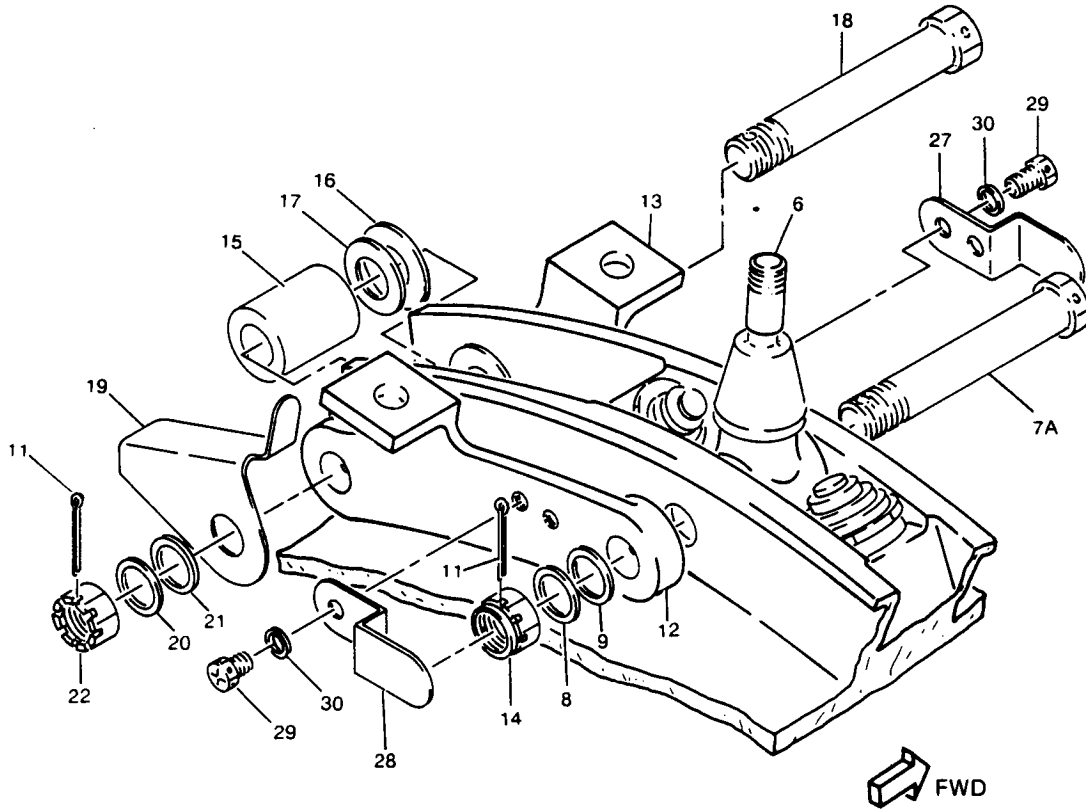


ENGINE NO.1
AFT MOUNT (-23 INSTL)
Engine Mount Cone Bolt Installation
Figure 1101 (Sheet 2)

OVERHAUL MANUAL

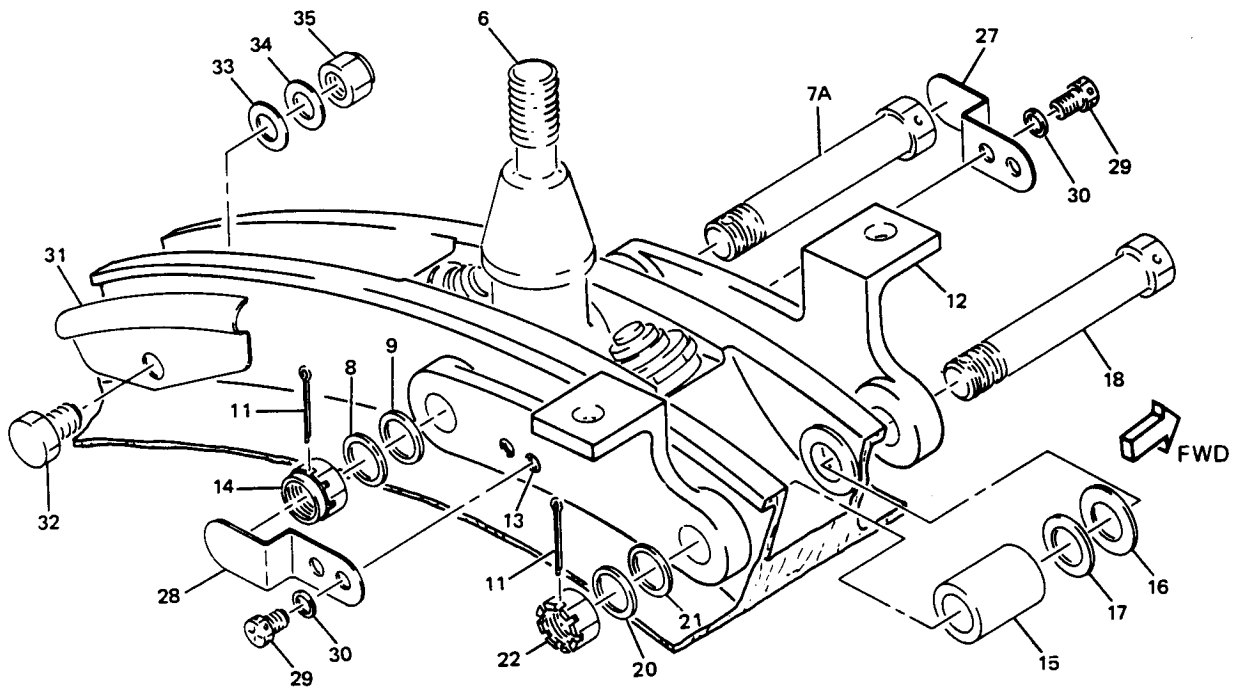


(D) ENGINE NO. 2
AFT MOUNT (-24 INSTL)
Engine Mount Cone Bolt Installation
Figure 1101 (Sheet 3)



ENGINE NO.1
AFT MOUNT (-29 INSTL)

(E)



ENGINE NO.2
AFT MOUNT (-30 INSTL)

(F)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1101-	65-19592-3		CONE BOLT INSTL, ENG MOUNT							A	RF
	65-19592-23		CONE BOLT INSTL, ENG MOUNT, ENG NO. 1 *[2]							B	RF
	65-19592-24		CONE BOLT INSTL, ENG MOUNT, ENG NO. 2 *[2]							C	RF
	65-19592-29		CONE BOLT INSTL, ENG MOUNT, ENG NO. 1 *[2]							D	RF
	65-19592-30		CONE BOLT INSTL, ENG MOUNT, ENG NO. 2 *[2]							E	RF
1	R18423-1		. CONE BOLT, FWD V13636 (BOEING 10-60517-39)(REPLD BY R18423-2)								
1	R18423-2		. CONE BOLT, FWD V13636 (BOEING 10-60517-43)(REPLS R18423-1)								2
1	R18423-49		. CONE BOLT, FWD V13636 (BOEING 10-60517-49)							D,E	2
2	NAS6712E64		. BOLT (PREF) *[00] (POST SB 737-71A1430)								2
2	BACB30GU12-64		. BOLT (OPT) *[00] (PRE SB 737-71A1430)								2
3	AN960C1216L		. WASHER								4
4	MS20002C12		. WASHER, COUNTERSUNK								2
5	BACN10BY-512		DELETED								
5	NAS1022A12		DELETED								
5	BACN10JC12		. NUT								2
6	R18424-1		. CONE BOLT, AFT V13636 (BOEING 10-60517-40) (REPLD BY R18424-2)							A	1
6	R18424-2		. CONE BOLT, AFT V13636 (BOEING 10-60517-44)(REPLS R18424-1)								1
7	66-13297-2		. BOLT (PRE SB 71-1069)							A	1
7A	66-13297-3		. BOLT							B-E	1
7A	66-13297-3		. BOLT (POST SB 71-1069)							A	
8	AN960C916L		. WASHER								1
9	AN960C916		. WASHER (PRE SB 71-1069)							A	2
9	AN960C916		. WASHER							B-E	*[1]
9	AN960C916		. WASHER (POST SB 71-1069)							A	*[1]
10	AN320C9		. NUT (PRE SB 71-1069)							A	1
11	AN381-4-16		. COTTER PIN (PRE SB 71-1069)							A	1
11	MS24665-370		. COTTER PIN							B-E	2
11	MS24665-370		. COTTER PIN (POST SB 71-1069)							A	2
12	69-72053-1		. FITTING, SIDE (PRE SB 71-1289)							B-E	1
12	69-72053-1		. FITTING, SIDE (POST SB 71-1069) (PRE SB 71-1289)							A	1
12	69-72053-3		. FITTING, SIDE (POST SB 71-1289)							B-E	1
12	69-72053-3		. FITTING, SIDE (POST SB 71-1289)							A	1
13	69-72053-2		. FITTING, SIDE (PRE SB 71-1289)							B-E	1
13	69-72053-2		. FITTING, SIDE (POST SB 71-1069) (PRE SB 71-1289)							A	1
13	69-72053-4		. FITTING, SIDE (POST SB 71-1289)							B-E	1
13	69-72053-4		. FITTING, SIDE (POST SB 71-1289)							A	1

Jul 1/02

71-03-01
Page 1102G

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1101-											
14	BACN10JD109AU									B-E	1
14	BACN10JD109AU									A	1
15	65-19592-25									B-E	1
15	65-19592-25									A	1
16	69B00412-6									B-E	AR
16	69B00412-6									A	AR
17	69B63073-501									B-E	AR
17	69B63073-501									A	AR
18	BACB30NF10D51									B-E	1
18	BACB30NF10D51									A	1
19	69-72056-2									BD	1
19	69-72056-2									A	1
19	69-72056-4									BD	1
19	69-72056-4									A	1
20	AN9601016L									B-E	1
20	AN9601016L									A	1
21	AN9601016									B-E	2
21	AN9601016									A	2
22	BACN10JD110AU									B-E	1
22	BACN10JD110AU									A	1
23	69-72057-1									BC	1
24	NAS6605-8									BC	2
25	AN960C516									BC	2
26	NAS1804-5									BC	2
27	69-72056-1									B-E	1
27	69-72056-1									A	1
28	69-72056-3									B-E	1
28	69-72056-3									A	1
29	NAS1801-3D6									B-E	4
29	NAS1801-3D6									A	4
30	AN960C10L									B-E	4
30	AN960C10L									A	4
31	69-29863-2									CE	1
32	NAS1105-3									CE	1
33	BACW10AK-WP5									CE	1
34	BACW10P-16S									CE	1
35	BACN10JC5									CE	1
-40	10-62188-1									DELETED	
-41	NAS6605-9									DELETED	
-42	AN960C516									DELETED	
-43	NAS1804-5									DELETED	
-44	BACB30LE5-79									DELETED	
-45	MS20002C5									DELETED	
46	NAS1804-5									BC	2

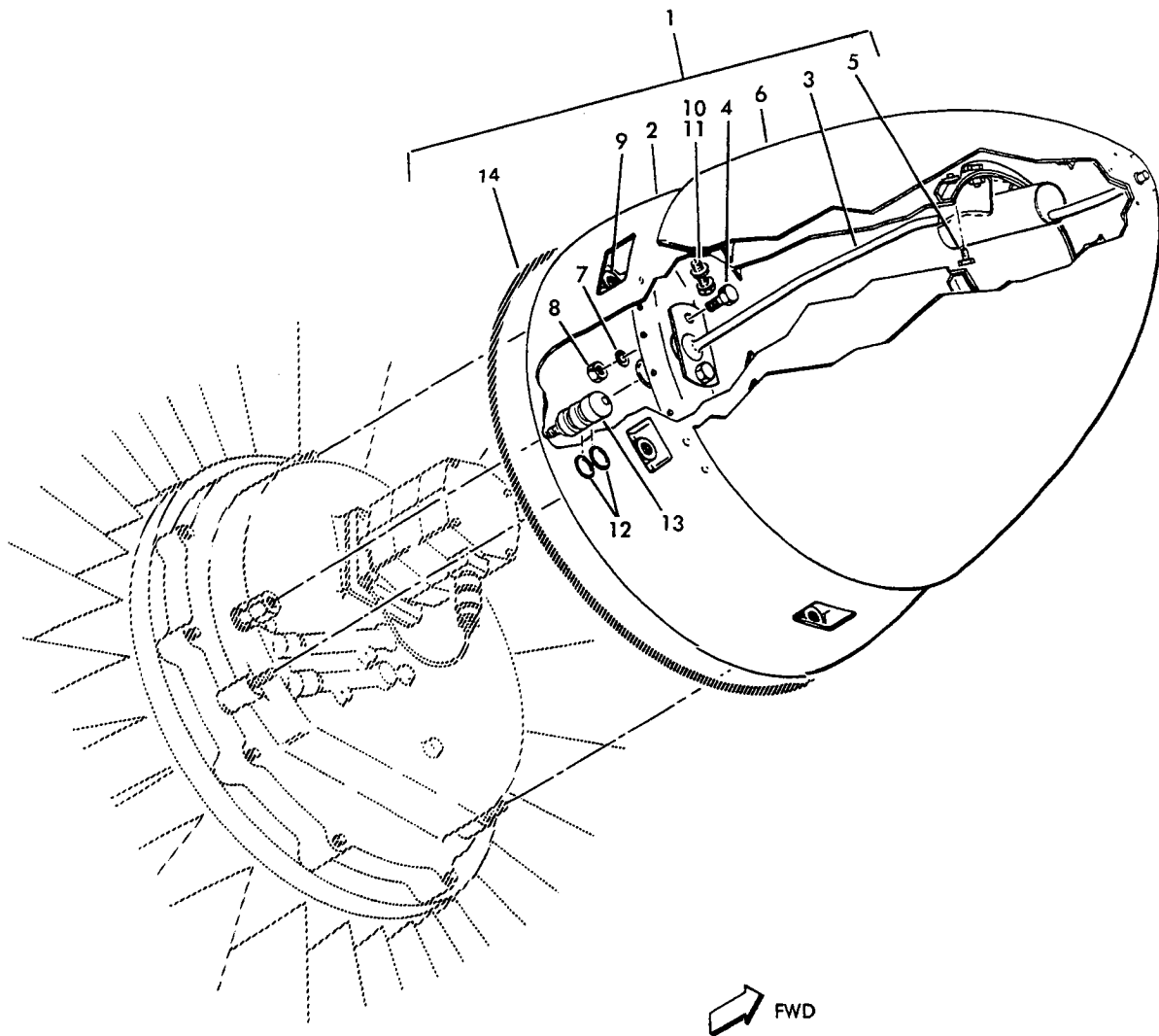
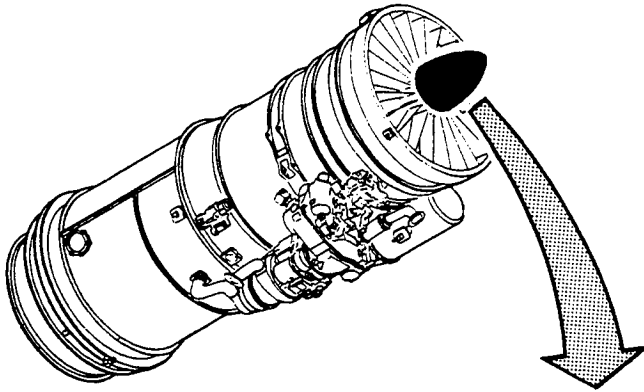
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY																																			
			1	2	3	4	5	6	7																																					
1101-47	AN960C516		.	W	A	S	H	E	R	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	2													
48	NAS6605-9		.	B	O	L	T	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	2															
49	84821CD-524		.	N	U	T	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	1																
50	AN960C516		.	W	A	S	H	E	R	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	1													
51	10-62188-2		.	S	P	A	C	E	R	,	C	R	U	S	H	A	B	L	E	(P	O	S	T	S	B		BC	1																	
52	BACB30LE5-79		.	B	O	L	T	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	1															
53	MS20002C5		.	W	A	S	H	E	R	,	C	O	U	N	T	E	R	S	I	N	K	(P	O	S	T	S	B		BC	1															
54	AN960C516		.	W	A	S	H	E	R	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B		BC	AR													
55	10-62188-1		.	L	O	A	D	L	I	M	I	T	E	R	(P	O	S	T	S	B	7	1	-	1	0	6	9)	(P	R	E	S	B	7	1	-	1	0	6	9)	*[3]	BC	1
56	NAS1804-6		.	N	U	T	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	2																						
57	AN960C516		.	W	A	S	H	E	R	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	2																			
58	NAS6606-10		.	B	O	L	T	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	2																					
59	BACN10HR5CD		.	N	U	T	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	1																						
60	69-76789-1		.	W	A	S	H	E	R	A	S	S	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	1																
61	69-76789-4		.	W	A	S	H	E	R	,	C	H	A	M	F	E	R	E	D	(P	O	S	T	S	B		BC	1																	
62	69-78715-1		.	B	O	L	T	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	1																					
63	65C26824-1		.	L	I	M	I	T	E	R	,	L	O	A	D	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	1													
64	MS24665-88		.	C	O	T	T	E	R	P	I	N	(P	O	S	T	S	B	7	1	-	1	2	8	9)	*[3]	BC	1																

*[00] LIMITED USAGE

*[1] INSTALL UP TO THREE WASHERS UNDER NUT. MINIMUM OF ONE WASHER REQUIRED UNDER NUT WITH COTTER PIN INSTALLED. NO WASHERS ALLOWED UNDER BOLTHEAD.

*[2] REFER TO OHM 71-23-01 FOR REPAIR INFORMATION.

*[3] THESE PARTS USED WITH CONE BOLT INSTALLATION 65-19592-3, -29, -30 AS INSTALLATION PARTS ONLY.



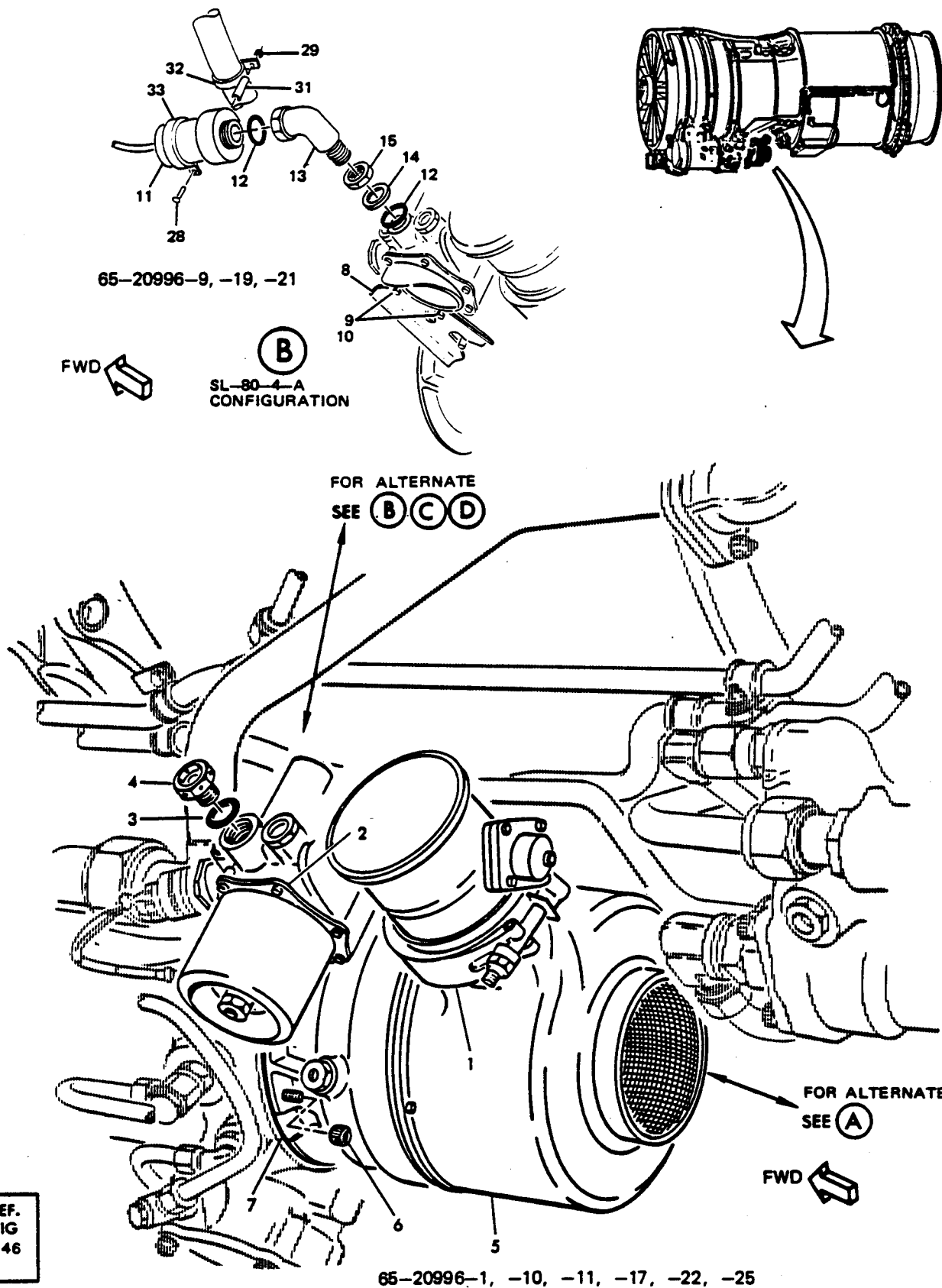
REF.
FIG
561

Engine Nose Dome Installation
Figure 1102

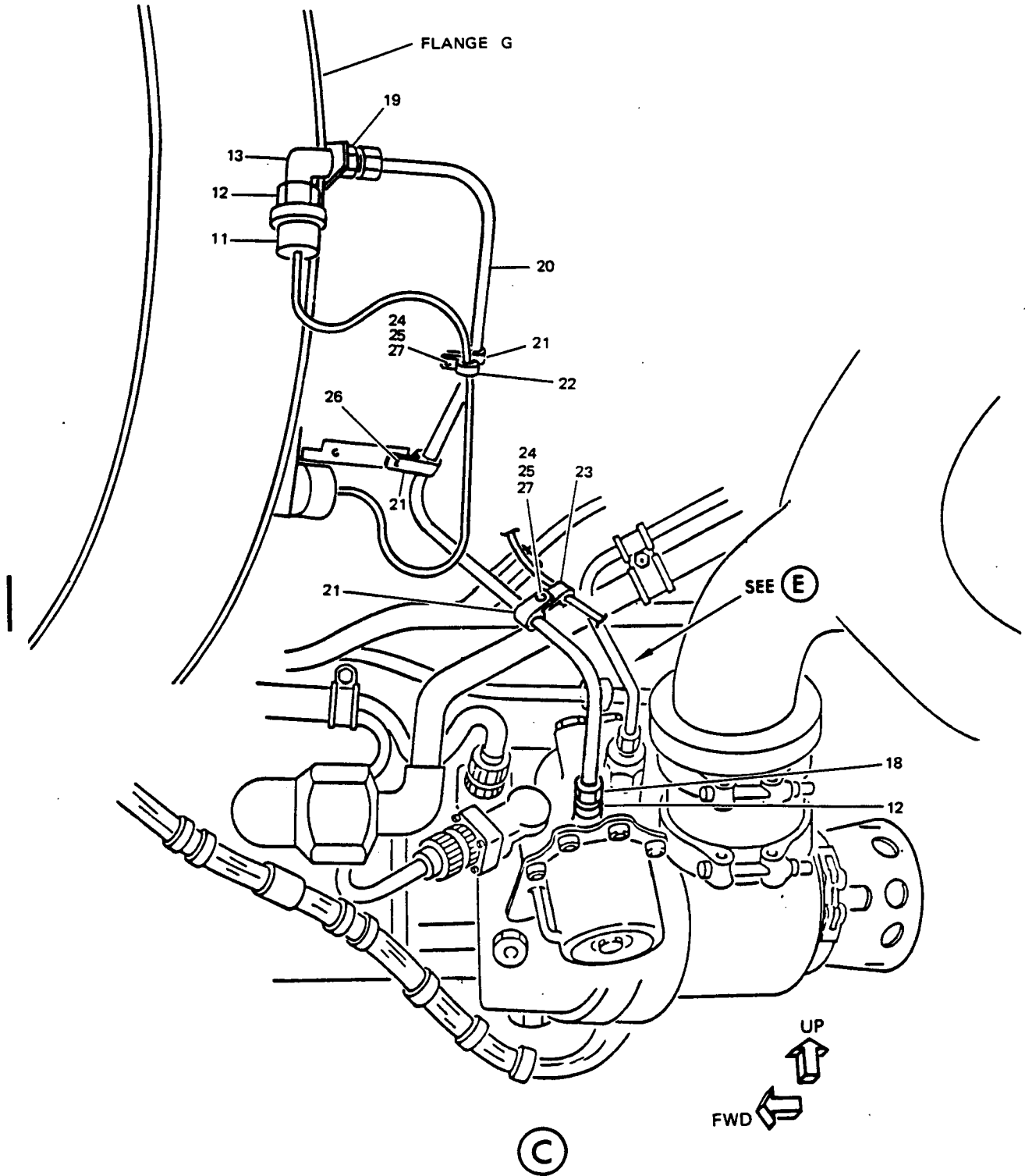
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1102-	65-19684-1		NOSE DOME INSTL, ENGINE *[1]*[4]								RF
1	65-19684-2		. NOSE DOME ASSY, ENGINE 1 *[2]*[5]								1
2	65-19684-17		. . NOSE CONE ASSY, ENGINE 1								
3	69-18491-3		. . . TUBE ASSY, PT2, NOSE DOME								1
4	NAS1103-4		. . BOLT								2
5	NAS1103-2		. . BOLT								2
6	65-19684-3		. . DOME ASSY								1
7	AN960D10		. . WASHER								2
8	NAS679A3W		. . NUT								2
9	BACN10NT5A		. . NUT, CAPTIVE								4
10	NAS1103-3		. . BOLT								16
11	AN960C10		. . WASHER								16
12	BACP11J113		. PACKING (PREF)								2
12	NAS1611-113		. PACKING								2
13	69-18713-800		. COUPLER, PT2, ENGINE NOSE DOME								1
14	*[3]		. PACKING								1

- *[1] PRE SB's 71-63 THRU 71-1065, 71-1067, 71-1071, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1125, 71-1198
- *[2] REFER TO 71-13-16 FOR OVERHAUL.
- *[3] REFER TO P&W IPC 72-23-00, FIG. 3 FOR PART NUMBER.
- *[4] REFER TO FIG. 1143 FOR POST SB 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1125, 71-1125R1, 71-1125R2, 71-1198 CONFIGURATIONS.
- *[5] PRE SB 71-1118, 71-1121, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151, 71-1179, 71-1278. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.

NOTE: Refer to Fig. 1143 for extended nose dome used with fat lip nose cowl.

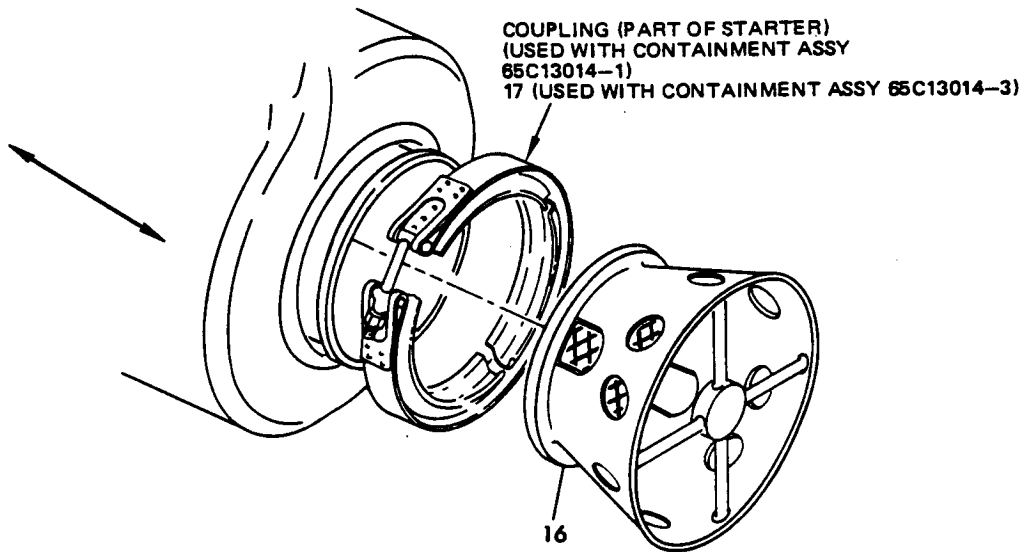


Starter and Valve Installation
Figure 1103 (Sheet 1)

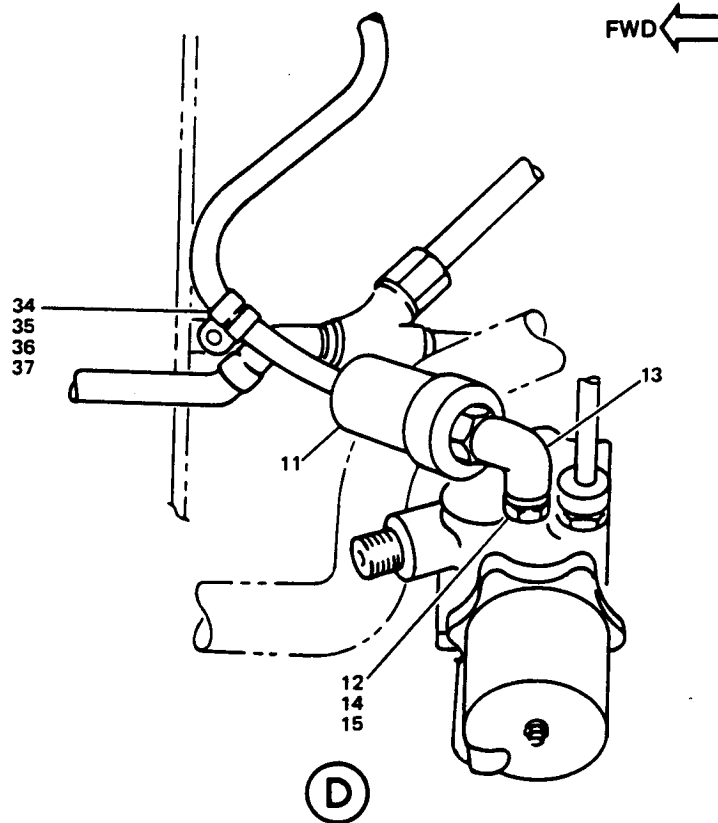


65-20996-23, -24

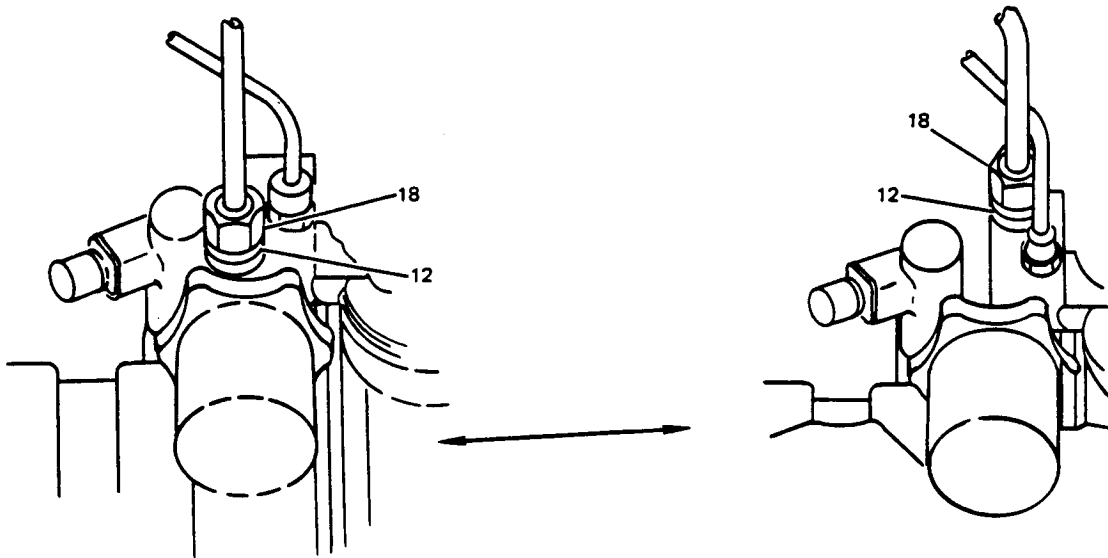
Starter and Valve Installation
Figure 1103 (Sheet 2)



(A)



Starter and Valve Installation
Figure 1103 (Sheet 3)



SB 80-1045 AND 80-1045R1
CONFIGURATION

(E)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1103-	65-20996-1									A	RF
	65-20996-9									B	RF
	65-20996-10									C	RF
	65-20996-11									D	RF
	65-20996-17									E	RF
	65-20996-19									F	RF
	65-20996-21									G	RF
	65-20996-22									H	RF
	65-20996-23									I	RF
	65-20996-24									J	RF
	65-20996-25									K	RF
1	BAC10DP300A										1
2	2730426									ABF-K	1
2	2680337									ACF-K	1
2	898172-1-1									AF-K	1
2	898172-3-1									AF-K	1
2	392688-5-1									AF-K	1
2	392688-1-1									AF-K	1
2	979078-1-1									D	1
2	979078-3-1									E	1
3	BACP11P-B6									ACDEHK	1
4	AN814-6DL										
4	AN814-6SL									ACDEHK	1
5	383152-16-1									ABCEF	1
										GIJK	
5	383152-1-2									D	1
5	383152-1-2									ABCEF	1
										GIJ	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY					
			1	2	3	4	5	6	7							
1103-5	383152-19-1		.	S	T	A	R	T	E	R	A	B	C	E	F	1
5	383342-1-1		.	S	T	A	R	T	E	R	A	H				1
6	BACN1OBL6L		.	N	U	T										5
7	AN4047-1		.	G	A	S	K	E	T			A	B	C	D	1
7	MS9136-01		.	G	A	S	K	E	T			F	G	I	J	1
8	66-20269-800		.	G	U	I	D	E	,	V	A	L	V	E	O	1
8	66-20269-2		.	G	U	I	D	E	,	V	A	L	V	E	O	1
8	66-20269-1		.	G	U	I	D	E	,	V	A	L	V	E	O	1
9	MS24673-3		.	S	C	R	E	W								2
10	BACW1OBP3APU		.	W	A	S	H	E	R	(U	S	E	W	I	2
10	AN960C10		.	W	A	S	H	E	R	(U	S	E	W	I	2
10	AN960PD10		.	W	A	S	H	E	R	(U	S	E	W	I	2
11	1G124		.	P	R	E	S	S	U	R	E	S	W	I	T	1
11	1G309		.	P	R	E	S	S	U	R	E	S	W	I	T	1
11	10-60927-3		.	P	R	E	S	S	U	R	E	S	W	I	T	1
12	BACP11PB6A		.	P	A	C	K	I	N	G	,	O	-	R	I	2
12	BACP11K6		.	P	A	C	K	I	N	G						1
13	2-01557		.	E	L	B	O	W	,	V	1	1	3	2	8	1
14	MS28777-6		.	W	A	S	H	E	R							1
15	BACN1ODU6		.	N	U	T										1
16	65C13014-1		.	C	O	N	T	A	I	N	E	M	E	N	T	1
16	65C13014-3		.	C	O	N	T	A	I	N	E	M	E	N	T	1
17	BACC10GY400		.	C	O	U	P	L	I	N	G	C	L	A	M	1
18	MS21902J6		.	U	N	I	O	N								1
19	BACN1ODR6J		.	N	U	T										1
20	65-58126-26		T	U	B	E	A	S	S							1
20	65-58126-27		T	U	B	E	A	S	S							1
21	BACC10HS06		C	L	A	M	P	(R	E	P	L	S	B	A	3
21	BACC10EP6		C	L	A	M	P									3
22	BACC10GE3C		C	L	A	M	P									1
23	BACC10GE4C		C	L	A	M	P									1
24	AN960C10L		W	A	S	H	E	R								2
25	NAS1802-3-6		S	C	R	E	W									2
26	NAS1802-3-10		S	C	R	E	W									1
27	BACN10JC3CM		N	U	T											2
28	NAS1801-3-24		S	C	R	E	W									1

737

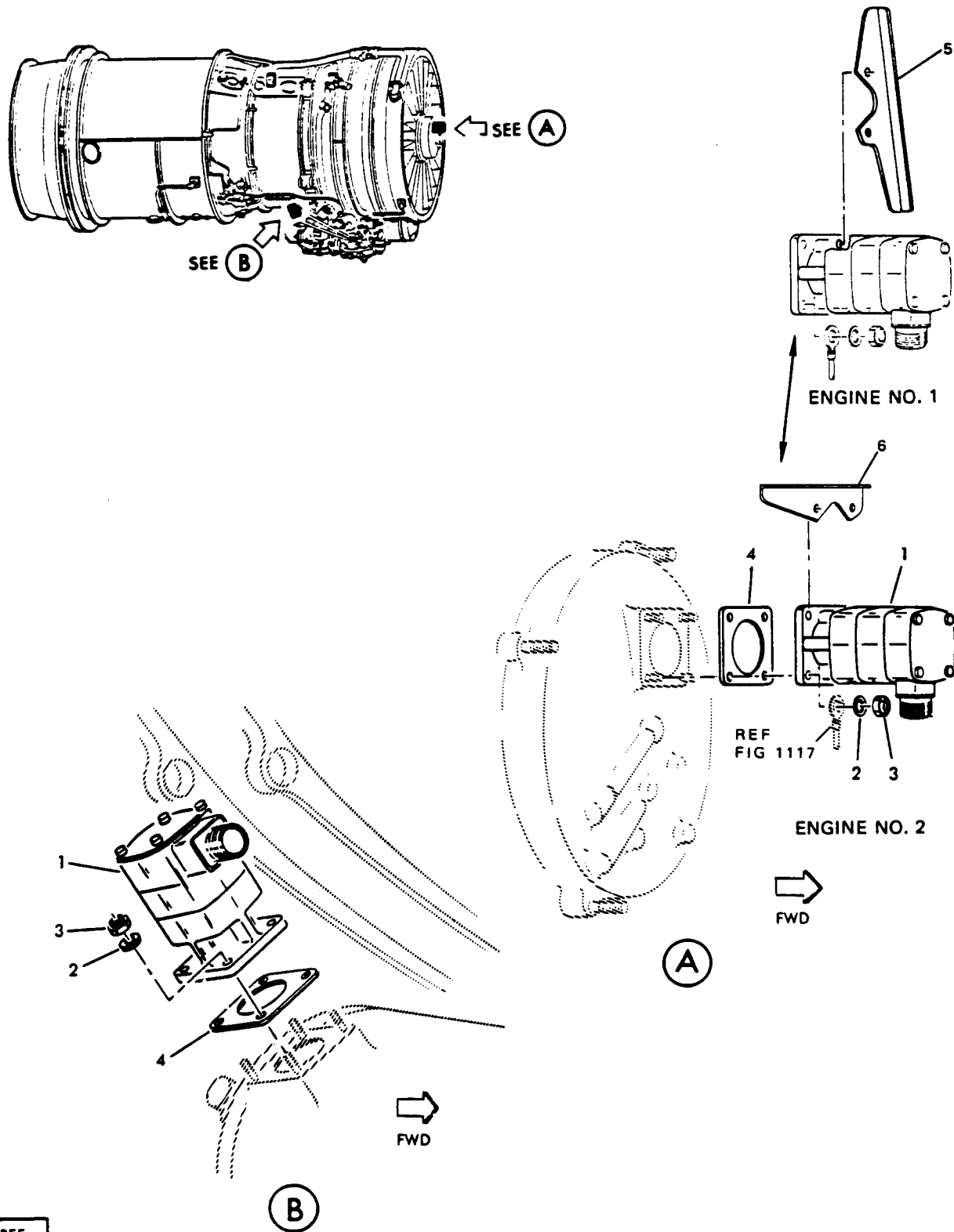
POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1103-29	BACN10JC3										1
-30	AN960D10										1
31	NAS43HT4-64										1
32	BACC10HS20										1
32	BACC10EP20										1
33	BACC10GE22C										1
34	BACC10GE4										1
35	NAS43HT3-80										1
36	NAS1801-3-30										1
37	AN960D10L										1

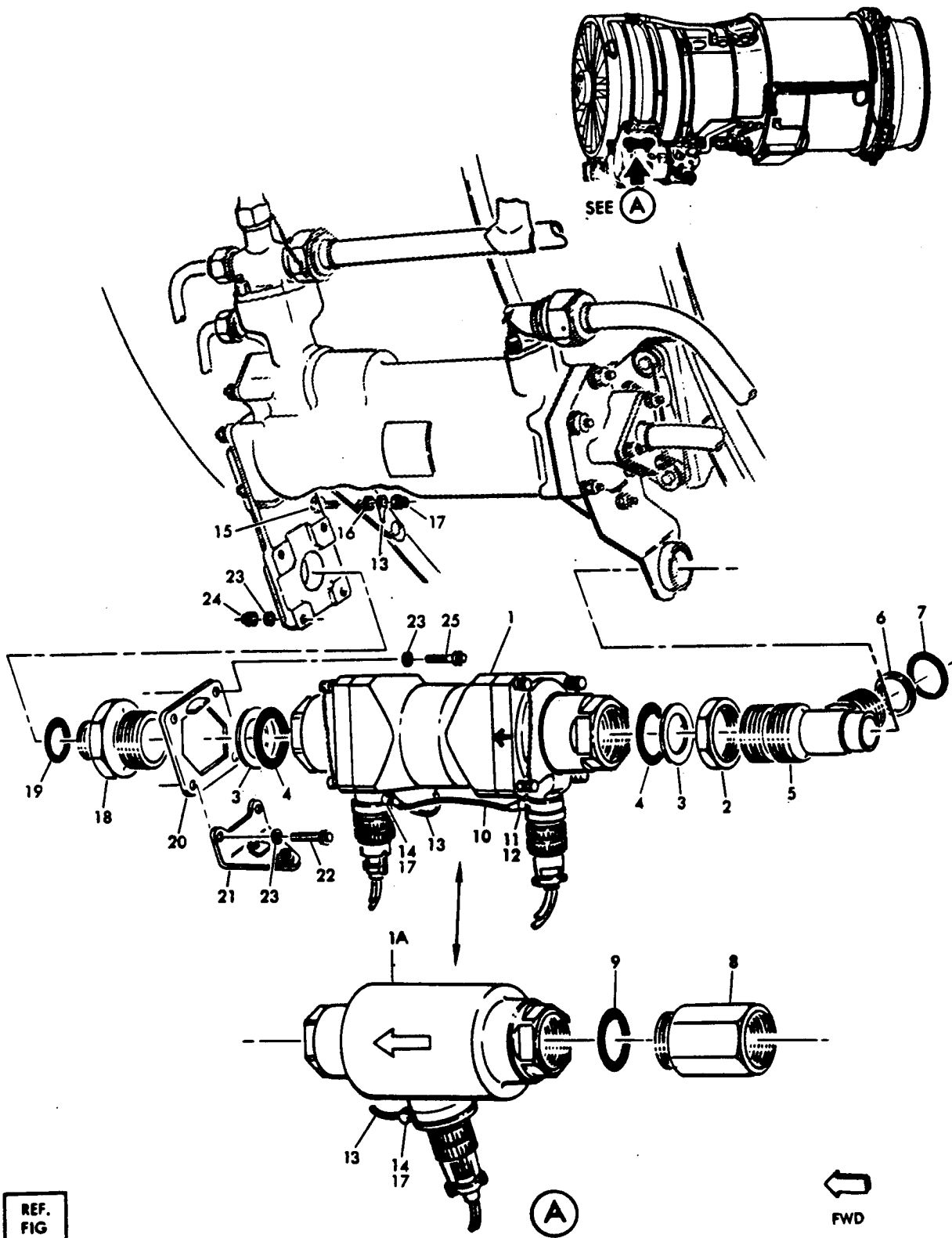
- *[1] LIMITED USAGE
- *[2] PRE SB 80-1004 THRU 80-1023, 80-1025 THRU 80-1039, 80-1044, 80-1051, 80-1052, 80-1055
- *[3] POST SB 80-1004 THRU 80-1008, 80-1019, 80-1055
- *[4] POST SB 80-1009 THRU 80-1017, 80-1020 THRU 80-1023, 80-1025 THRU 80-1039, 80-1044, 80-1047, 80-1048
- *[5] USED WITH PARKER (V92003) START VALVE
- *[6] USED WITH AIRESEARCH (V99193) START VALVE
- *[7] POST SB 80-1045, 80-1045R1, 80-1050, 80-1055
- *[8] LIMITED USAGE. NOT PART OF STARTER AND VALVE INSTALLATION
- *[9] INTERIM START VALVE OPEN INDICATION SYSTEM CONFIGURATION PER SL-80-4-A
- *[10] PRE SB 80-1045, 80-1045R1
- *[11] POST SB 80-1051, 80-1052, 80-1057



REF.
 FIG
 531
 560

N1 and N2 Tachometer Generator Installation
 Figure 1104

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1104-	65-21950-2		GENERATOR INSTL - N1 AND N2 TACHOMETER							A	RF
	65-21950-3		GENERATOR INSTL - N1 AND N2 TACHOMETER							B	RF
	65-21950-5		GENERATOR INSTL - N1 AND N2 TACHOMETER (ENG. NO. 1)							C	RF
	65-21950-6		GENERATOR INSTL - N1 AND N2 TACHOMETER (ENG. NO. 2)							D	RF
	65-21950-7		GENERATOR INSTL - N1 AND N2 TACHOMETER (NO. 1)							E	RF
	65-21950-8		GENERATOR INSTL - N1 AND N2 TACHOMETER (NO. 2)							F	RF
1	65-21950-9		GENERATOR INSTL - N1 AND N2 TACHOMETER (ENG. NO. 1)							G	RF
1	65-21950-10		GENERATOR INSTL - N1 AND N2 TACHOMETER (ENG. NO. 2)							H	RF
1	B5069110302-0		. TACHOMETER GENERATOR, V89944							AEF	2
1	2CM9ABY-7		. TACHOMETER GENERATOR, V97424							BCD	2
1	2CM15AAB1		. TACHOMETER GENERATOR, V97424							GH	2
2	AN960PD416L		. WASHER								8
3	NAS679A4W		. NUT								8
4	AN4045-1		. GASKET								2
5	65C13050-1		. BRACKET, INTERFERENCE - ENG NO. 1							CE	1
5	65C13050-1		. BRACKET, INTERFERENCE - ENG NO. 1 (POST SB'S 71-1115, 71-1116, 71-1093, 71-1101R1, 71-1151)								1
6	65C13050-2		. BRACKET, INTERFERENCE - ENG NO. 2							DF	1
6	65C13050-2		. BRACKET, INTERFERENCE - ENG NO. 2 (POST SB'S 71-1115, 71-1116, 71-1093, 71-1101R1, 71-1151)								1

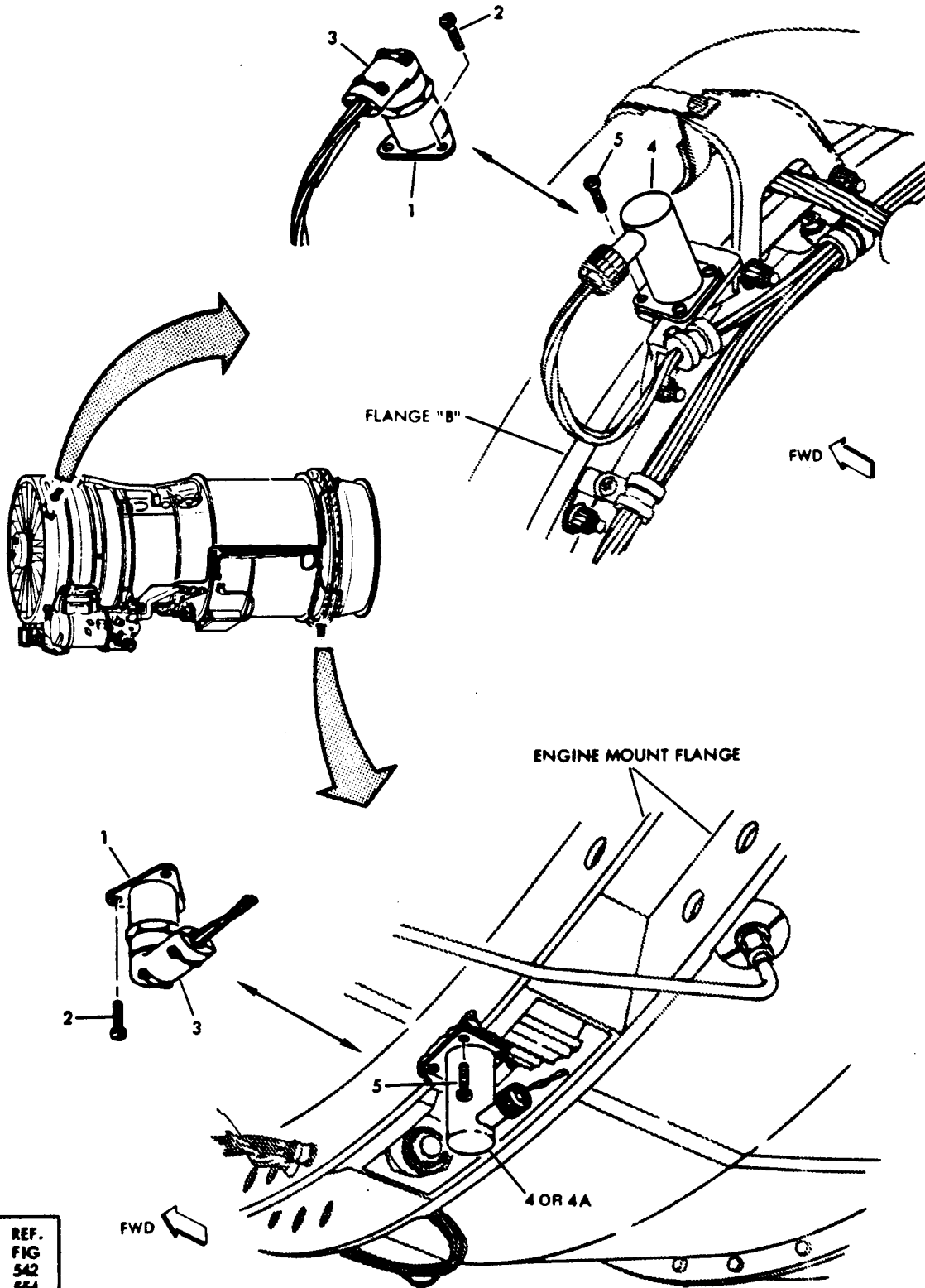


Fuel Flowmeter Installation
 Figure 1105

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1105-	65-22249-1		METER INSTL, FUEL FLOW							A	RF
	65-22249-2		METER INSTL, FUEL FLOW							B	RF
	65-22249-3		METER INSTL, FUEL FLOW							C	RF
1	8TJ50GCA		. TRANSMITTER ASSY, FUEL FLOWMETER (V97424)(BOEING 10-60507-6)*[1]							A	1
1	8TJ50GBZ		. TRANSMITTER ASSY, FUEL FLOWMETER (V97424)(BOEING 10-60507-1)*[1]							A	1
1	8TJ50GBZ		. TRANSMITTER ASSY, FUEL FLOWMETER (V97424)(BOEING 10-60507-1)							B	1
1A	8TJ85GHA		. TRANSMITTER ASSY, FUEL FLOWMETER (V97424)(BOEING 10-60507-14)*[1]							C	1
1A	8TJ85GAZ		. TRANSMITTER ASSY, FUEL FLOWMETER (V97424)(BOEING 10-60507-17)*[1]							C	1
2	MS9099-20		. NUT								1
3	MS9058-20		. RING, RETAINER								2
4	MS29512-20		. PACKING (PREF)								2
4	MS9020-20		. PACKING (OPT)								2
5	773089		. CONNECTOR TUBE (V77445)(OPT 447887, 762712, 773083)								1
5	447887		. CONNECTOR TUBE (V77445) (OPT 773089, 762712, 773083)								1
5	762712		. CONNECTOR TUBE (V77445) (OPT. 773089, 773083, 447887)								
5	773083		. CONNECTOR TUBE (V77445) PREF.								
6	354117		. RING, RETAINER, (V77445)								1
7	226195		. PACKING, SILASTIC, (V77445)								1
8	69-35244-1		. ADAPTER, FUEL FLOWMETER							C	1
9	MS29512-20		. PACKING, (PREF)							C	1
9	MS9020-20		. PACKING, (OPT)							C	1
10	BACJ40A20-5		. JUMPER ASSY							AB	1
11	BACS12CB3-6		. SCREW							AB	1
12	NAS679A3W		. NUT							AB	1
13	BACJ40A20-9		. JUMPER ASSY								1
14	BACS12CB3-7		. SCREW								1
15	BACS12CB3-9		. SCREW								1
16	AN960D10		. WASHER								1
17	NAS679A3W		. NUT								2

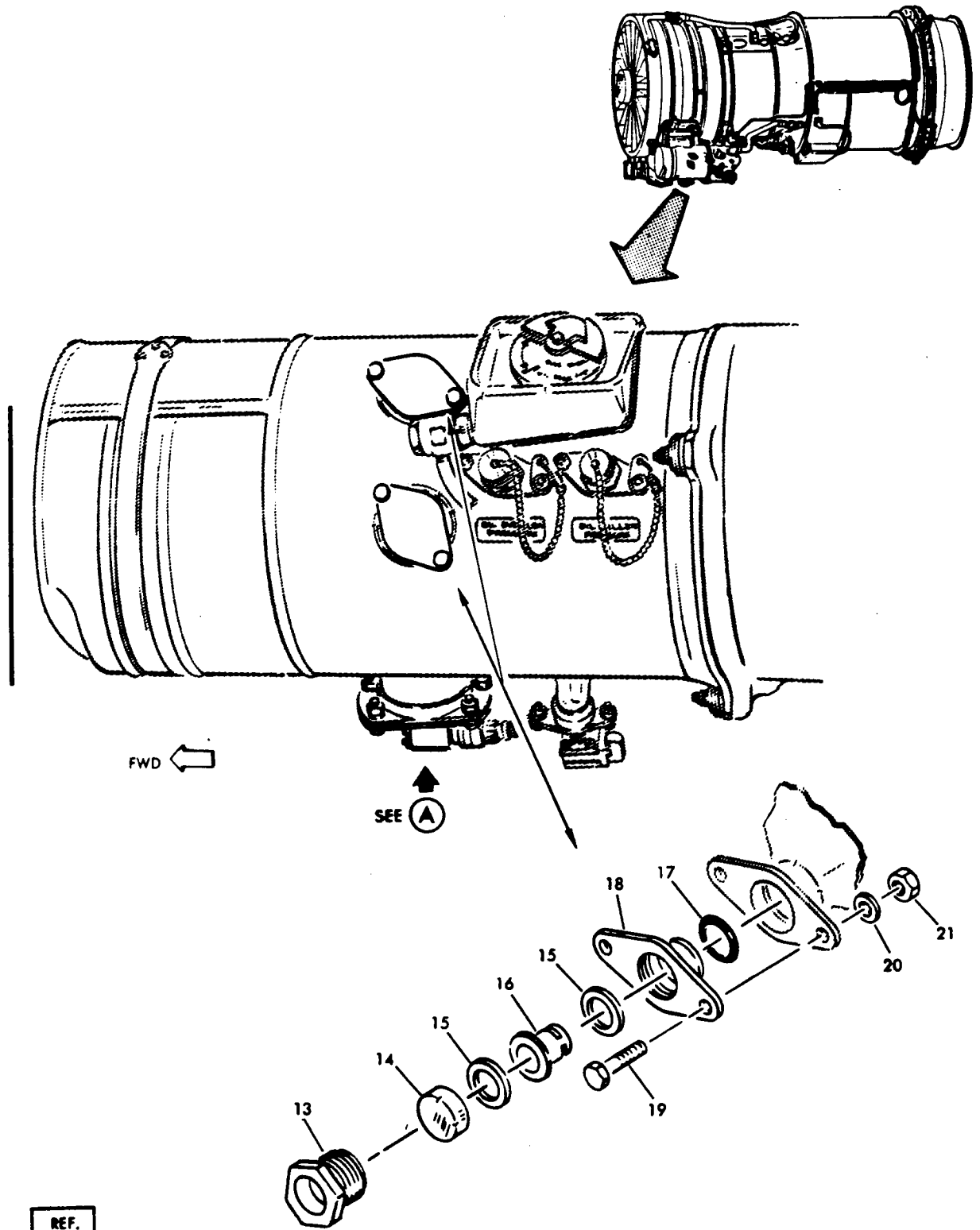
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1105-18	773086										1
18	447884										1
19	MS29513-21										1
19	MS9021-021										1
20	447885										1
21	66-15227-5										1
22	MS9147-18										3
23	AN122582										12
24	372873										6
25	MS9147-22										1

*[1] LIMITED USAGE



Engine Vibration Pickup Installation
Figure 1106

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1106-	65-22454-3									A	RF
	65-22454-7									A	RF
	65-22454-8									B	RF
	65-22454-9									C	RF
	65-22454-13									D	RF
	65-22454-14									E	RF
	65-22454-15									F	RF
	65-22454-16									G	RF
	65-22454-20									H	RF
	65-22454-24									I	RF
	65-22454-21									J	RF
	65-22454-25									K	RF
1	1510376-2									AJ	2
2	AN500AD8-7									AJ	6
3	1784340-10									A	2
4	4-125-0001									BC	2
4	4-126-0001									EH	2
4	MODEL-750									DG	2
4	WL-800									D	2
4	MODEL 9145									D	2
4	WL-750									F	2
4	1784340-60									J	2
4	144-904-000-01									K	2
4	144-904-000-04									K	2
4	144-904-000-05									K	2
4A	4-125-0001									I	1
5	AN500AD6-8									B-H	8
5	AN500AD6-8									I	4



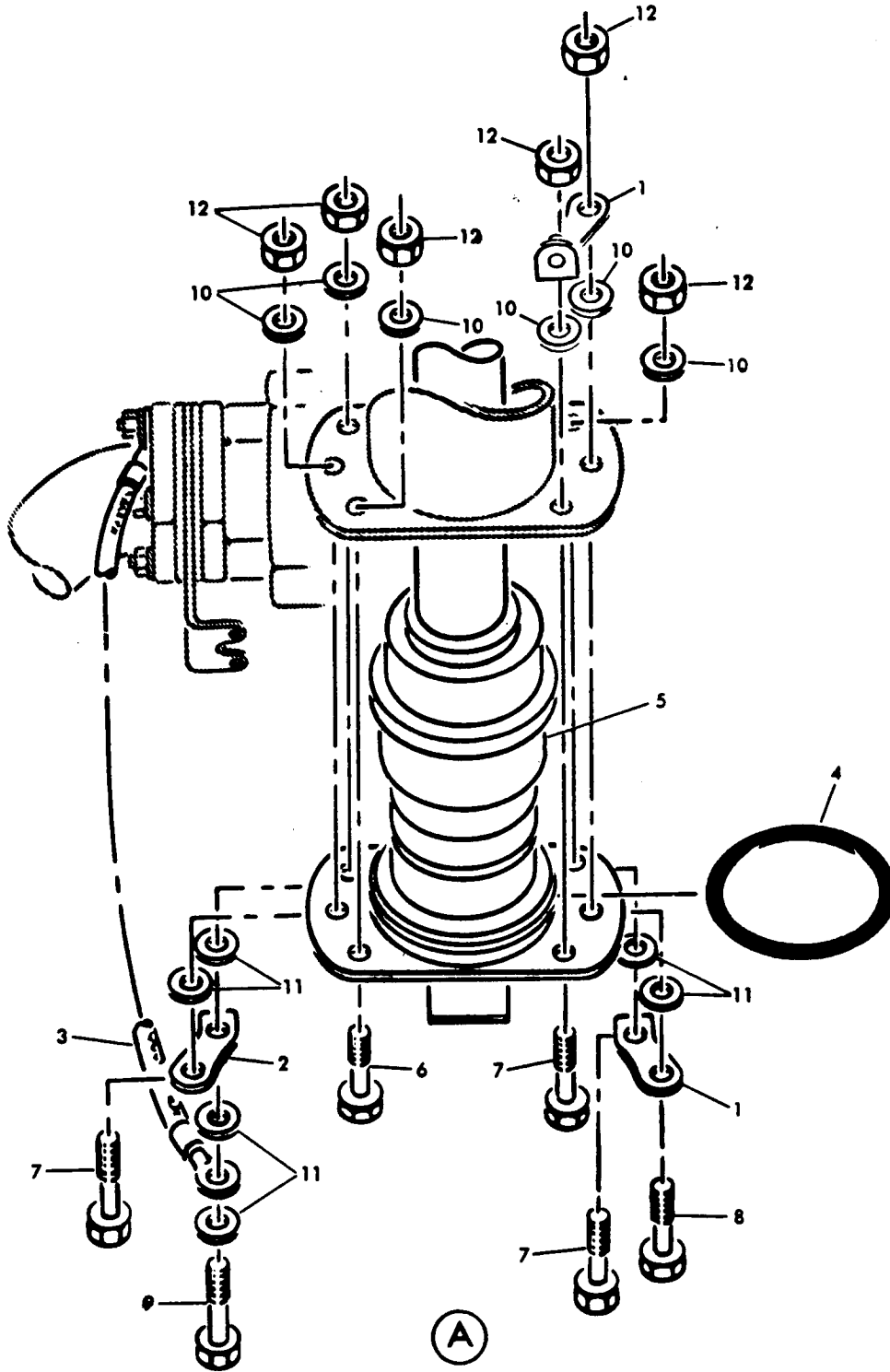
REF.
FIG
540

SIGHT GLASS INSTALLATION

Engine Oil Quantity Tank Unit Installation
Figure 1107 (Sheet 1)

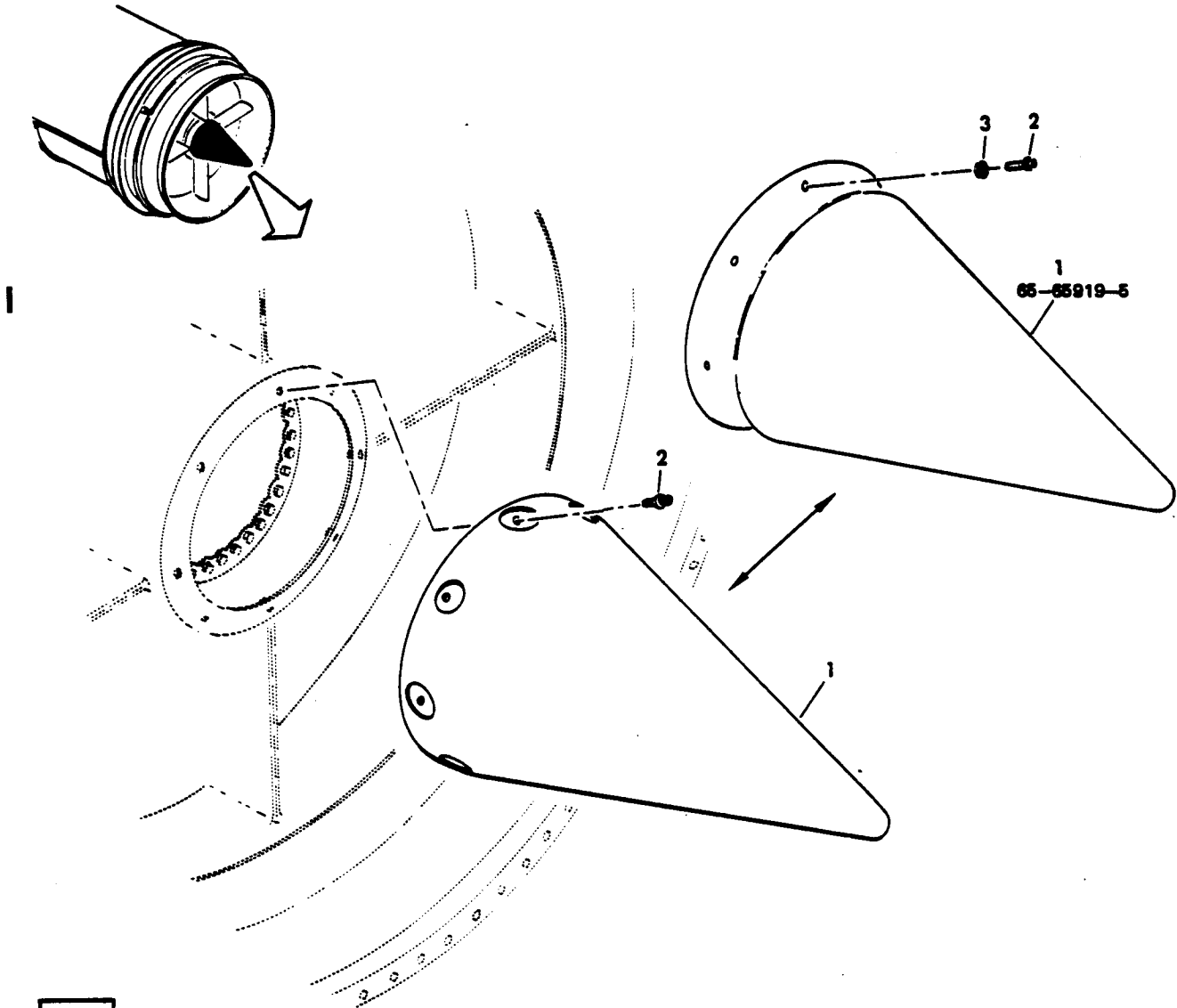
Sep 10/71

71-03-01
Page 1108A



Engine Oil Quantity Tank Unit Installation
Figure 1107 (Sheet 2)

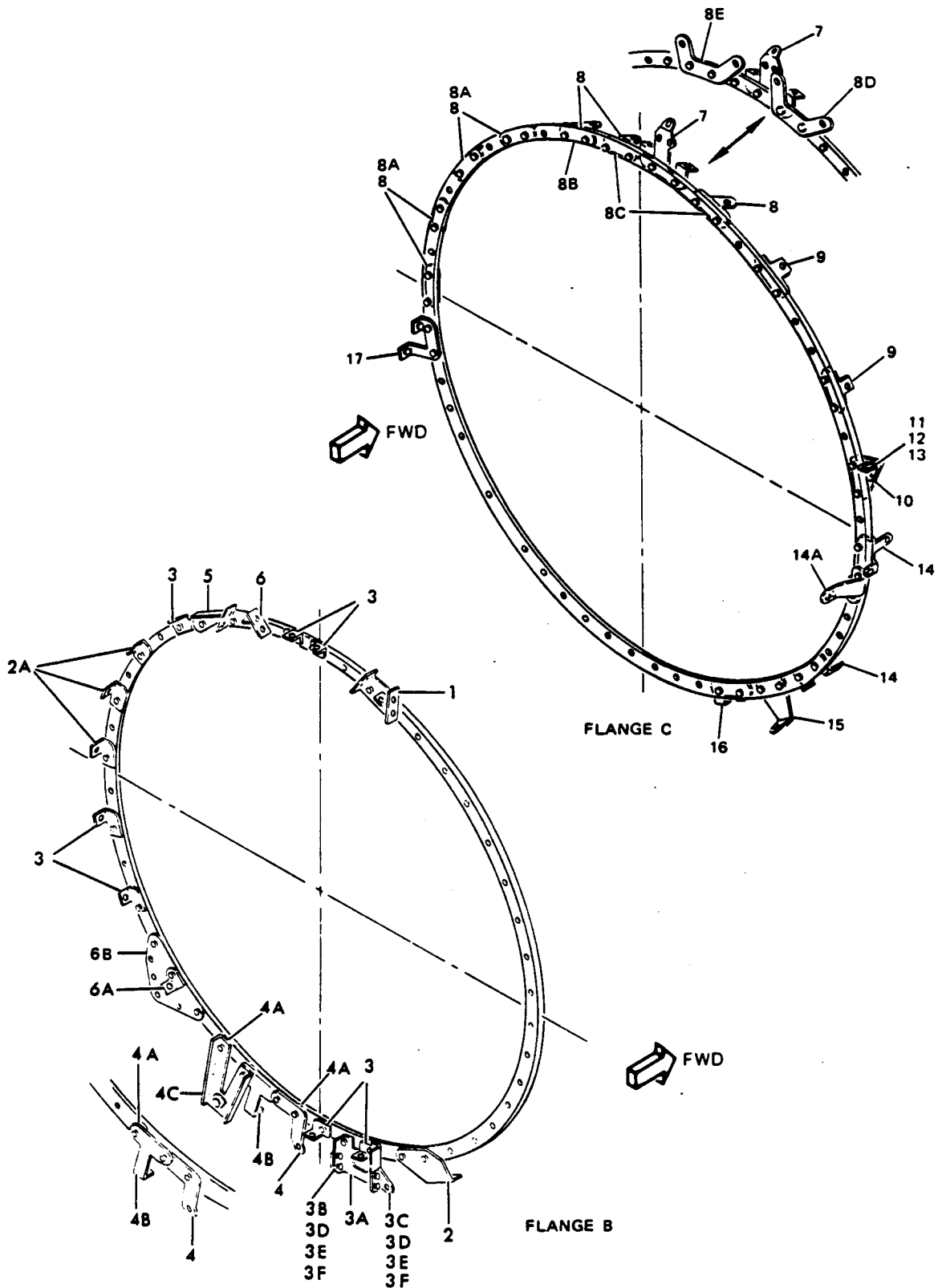
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1107	65-25736-2		TANK UNIT INSTL, OIL QUANTITY								RF
1	69-45804-2		. BRACKET								2
2	69-45804-3		. BRACKET								1
3	BACJ40K7A6A6		. JUMPER, BONDING								1
4	BACP11P228		. PACKING								1
5	10-60722-4		. TANK UNIT								1
			APPROVED PARTS ARE:								
			391012-01001, V77932								
			391080-08096, V77932								
6	NAS1104-4		. BOLT								1
7	NAS1104-6		. BOLT								3
8	NAS1104-7		. BOLT								1
9	NAS1104-8		. BOLT								1
10	AN960C416		. WASHER								6
11	AN960C416L		. WASHER								6
12	NAS679C4W		. NUT								6
	65-25736-3		SIGHT GLASS INSTL, ENGINE OIL QUANTITY								RF
13	BACC14B1CH		. CAP								2
14	BACG21A10		. GLASS, SIGHT								2
15	BACW10H101		. WASHER								4
16	3-58452		. REFLECTOR								2
17	BACP11P26		. PACKING								2
18	69-31653-1		. FITTING, SIGHT GAGE, (PREF)								
18	66-15330-1		. FITTING, SIGHT GAGE, (OPT)								
19	NAS1304-2HW		. BOLT								4
20	AN960C416L		. WASHER								4
21	NAS679C4W		. NUT								4



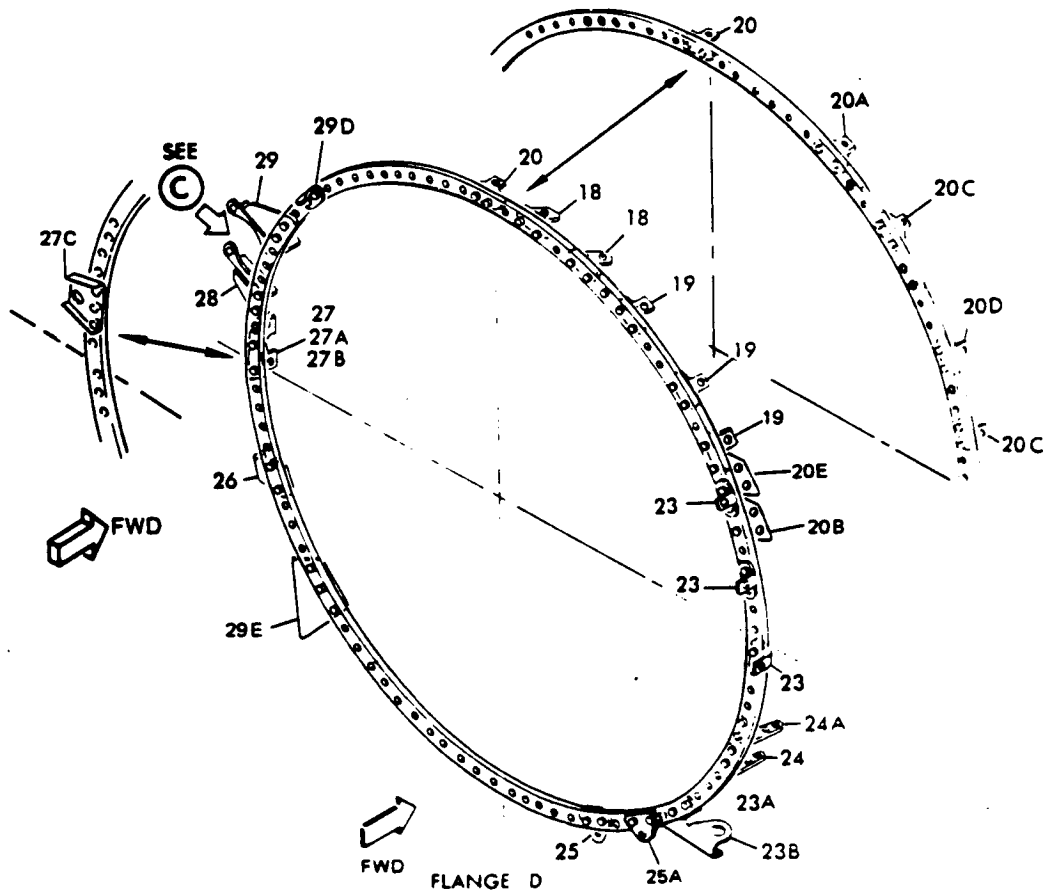
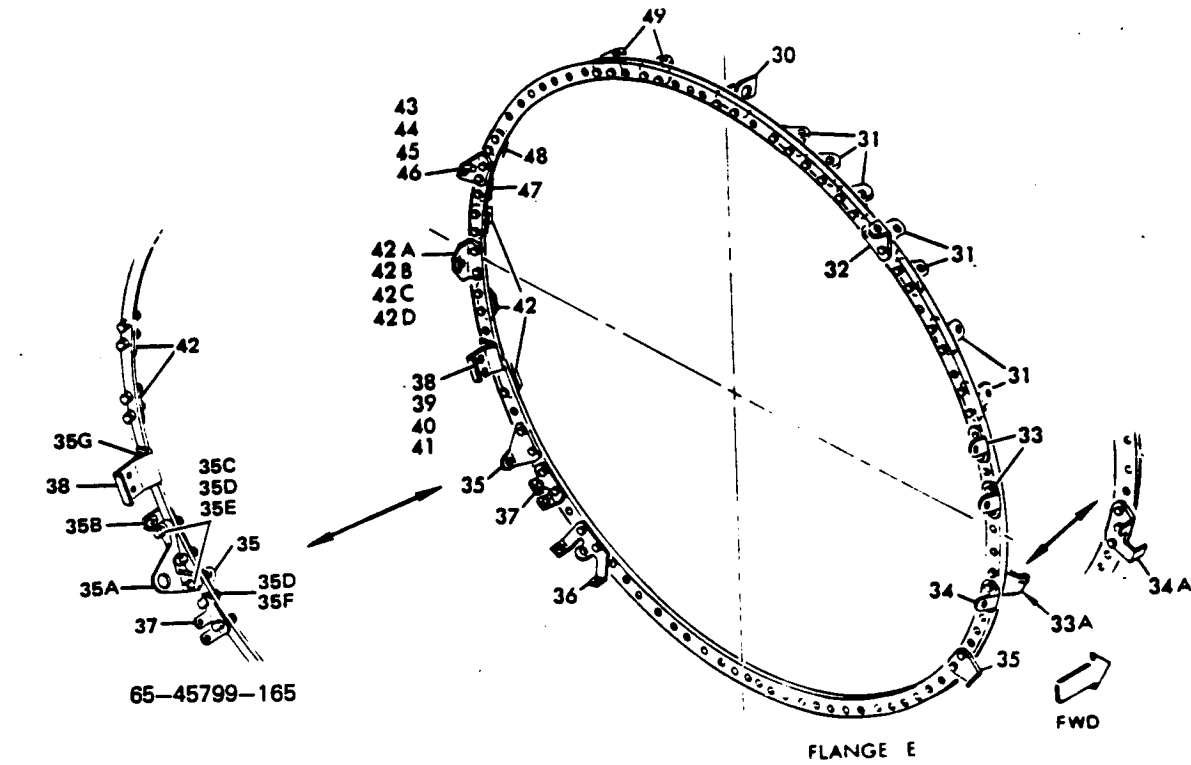
REF.
FIG
556

Tail Plug Installation
Figure 1108

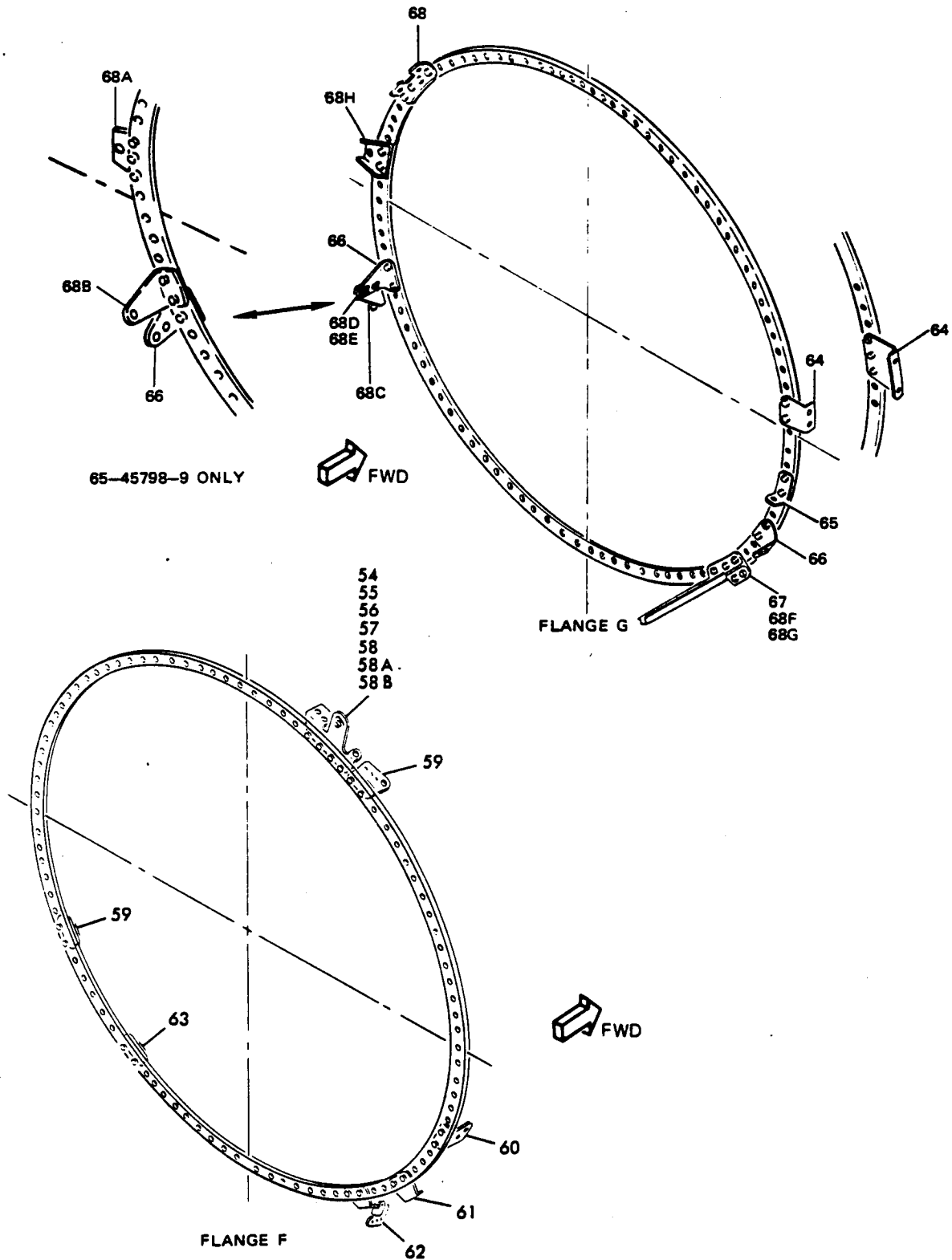
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1108-	65-27808-1									A	RF
	65-65919-4									B	RF
1	65-27808-800									A	1
1	65-27808-14									A	1
1	65-65919-5									B	1
2	MS9034-10									B	8
2	MS9034-08									A	8
3	AN960C416									B	8

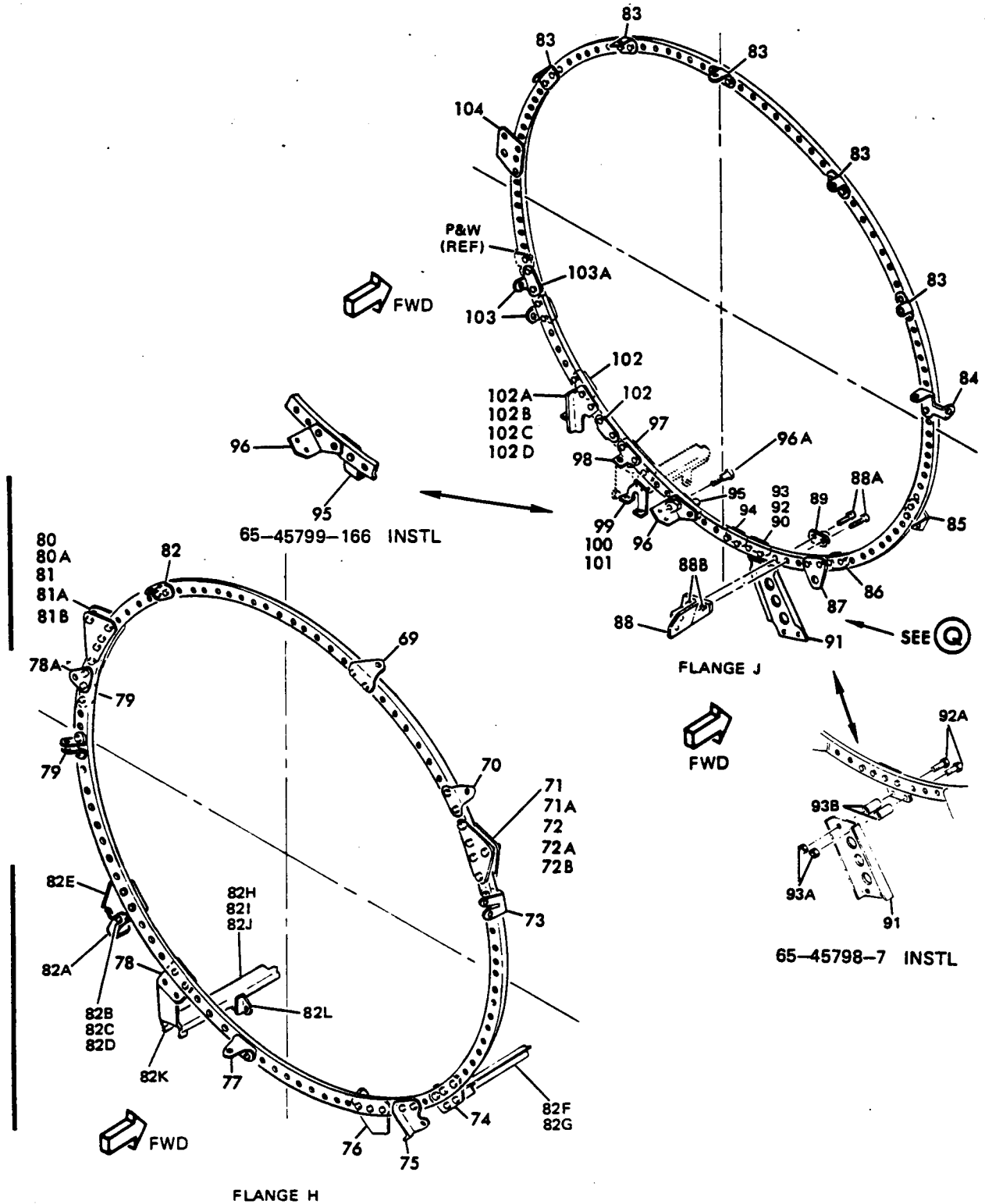


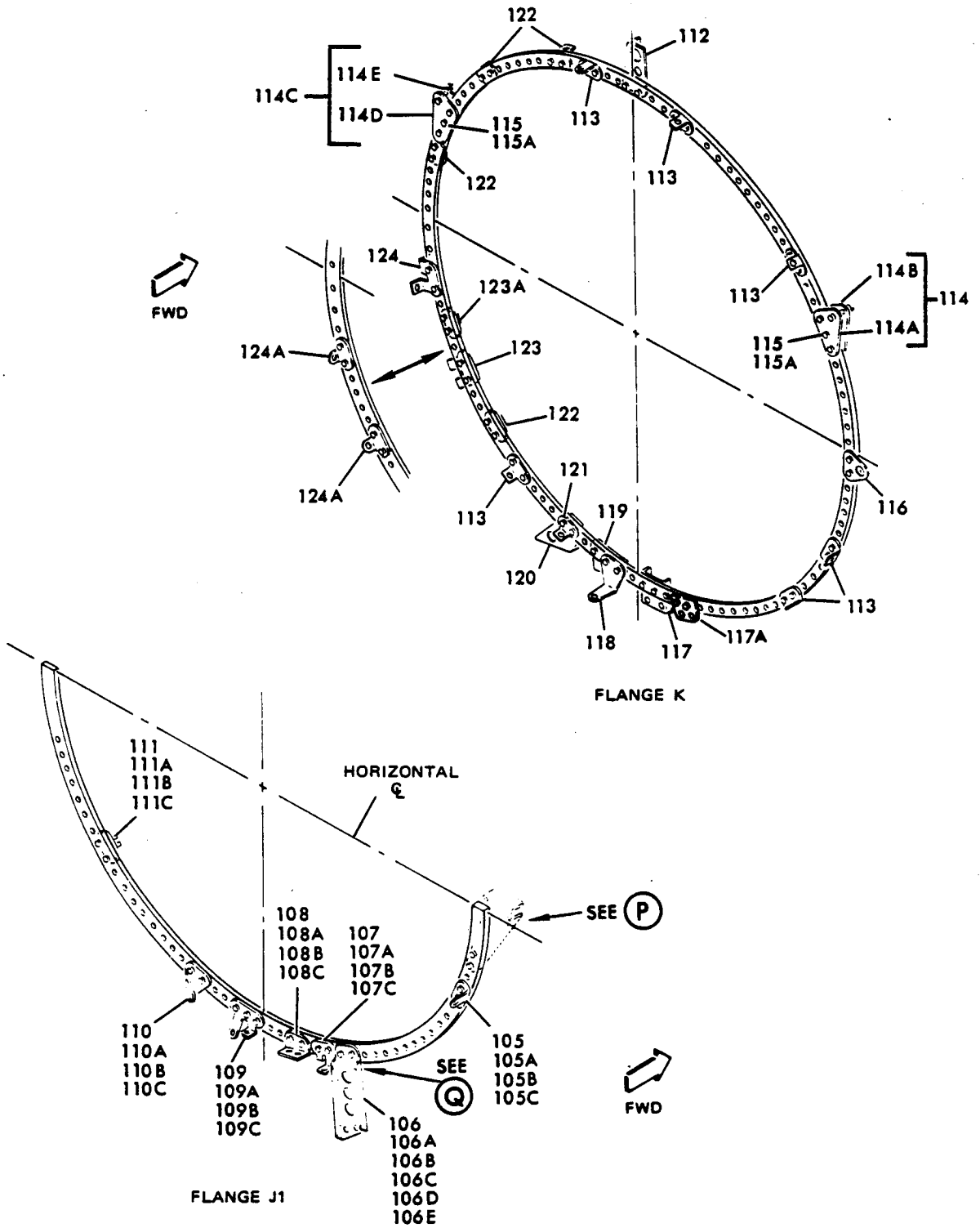
OVERHAUL MANUAL



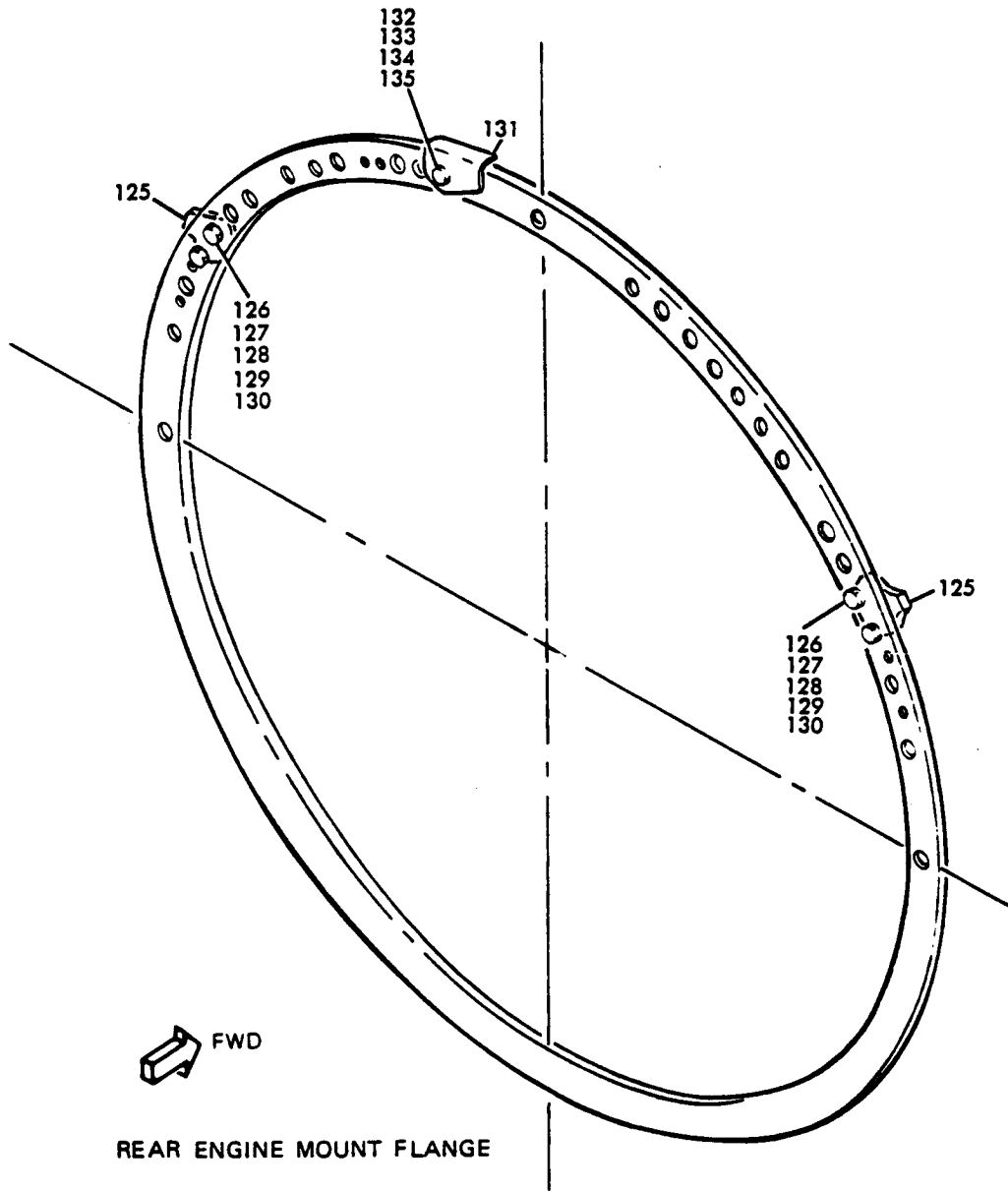
Bracket Installation
Figure 1109 (Sheet 2)

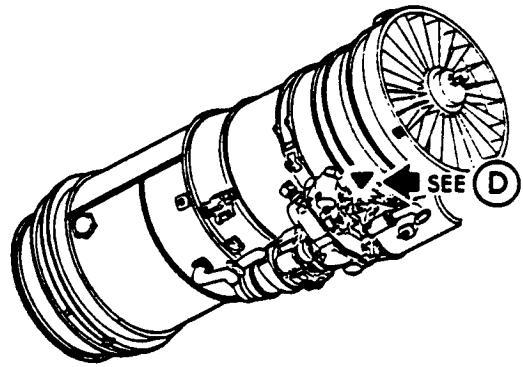
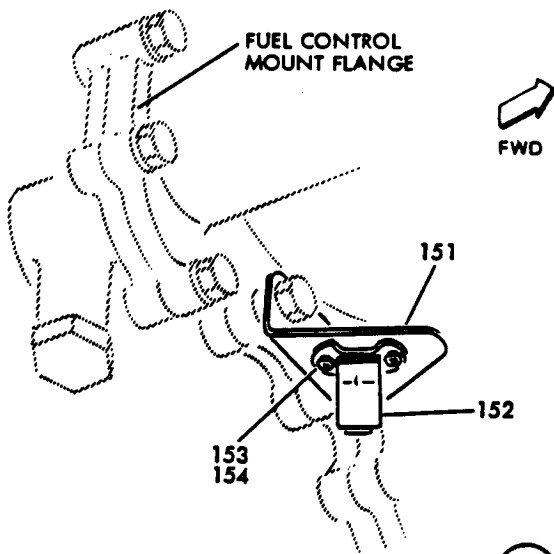
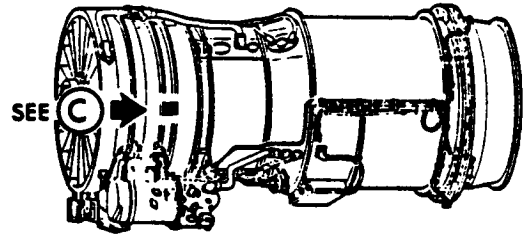
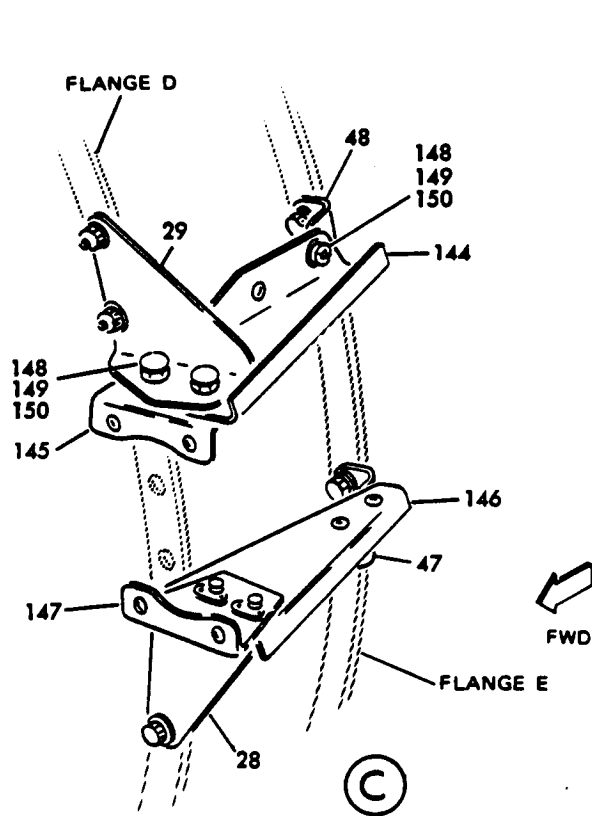




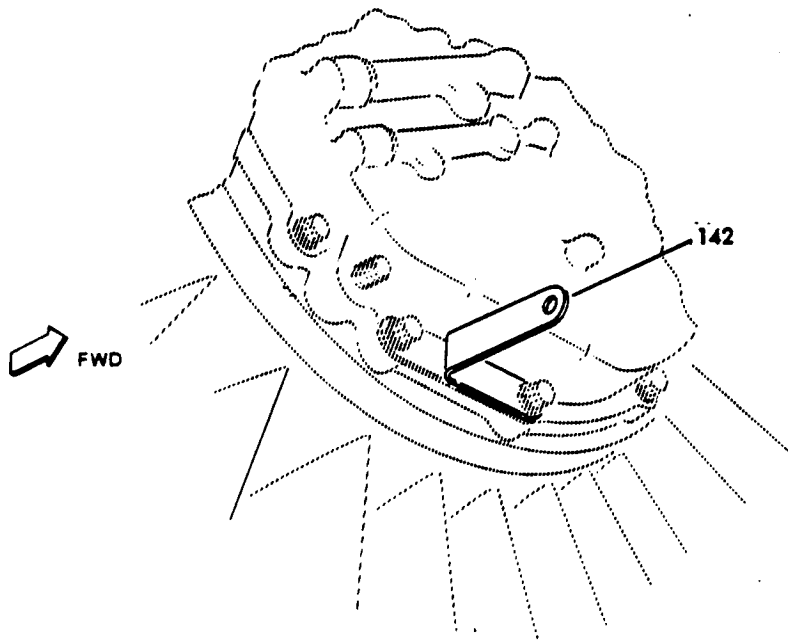
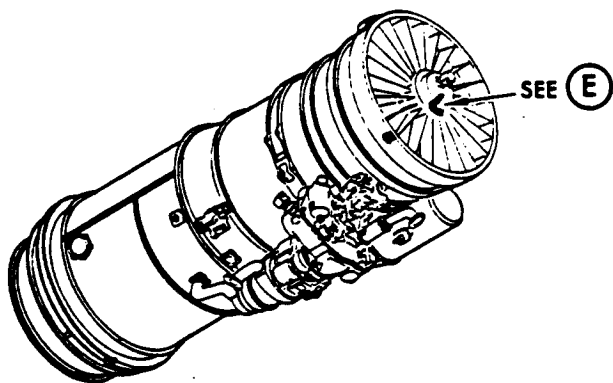


Bracket Installation
Figure 1109 (Sheet 5)



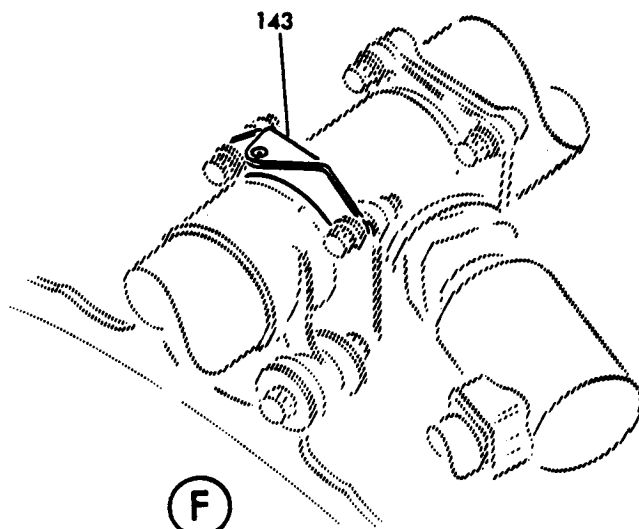
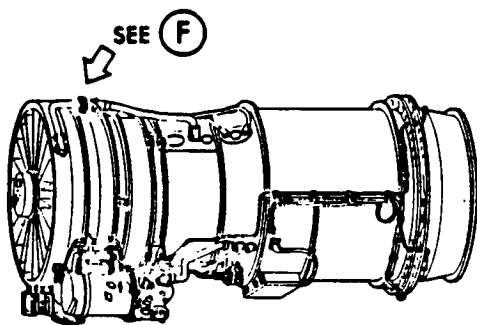


D



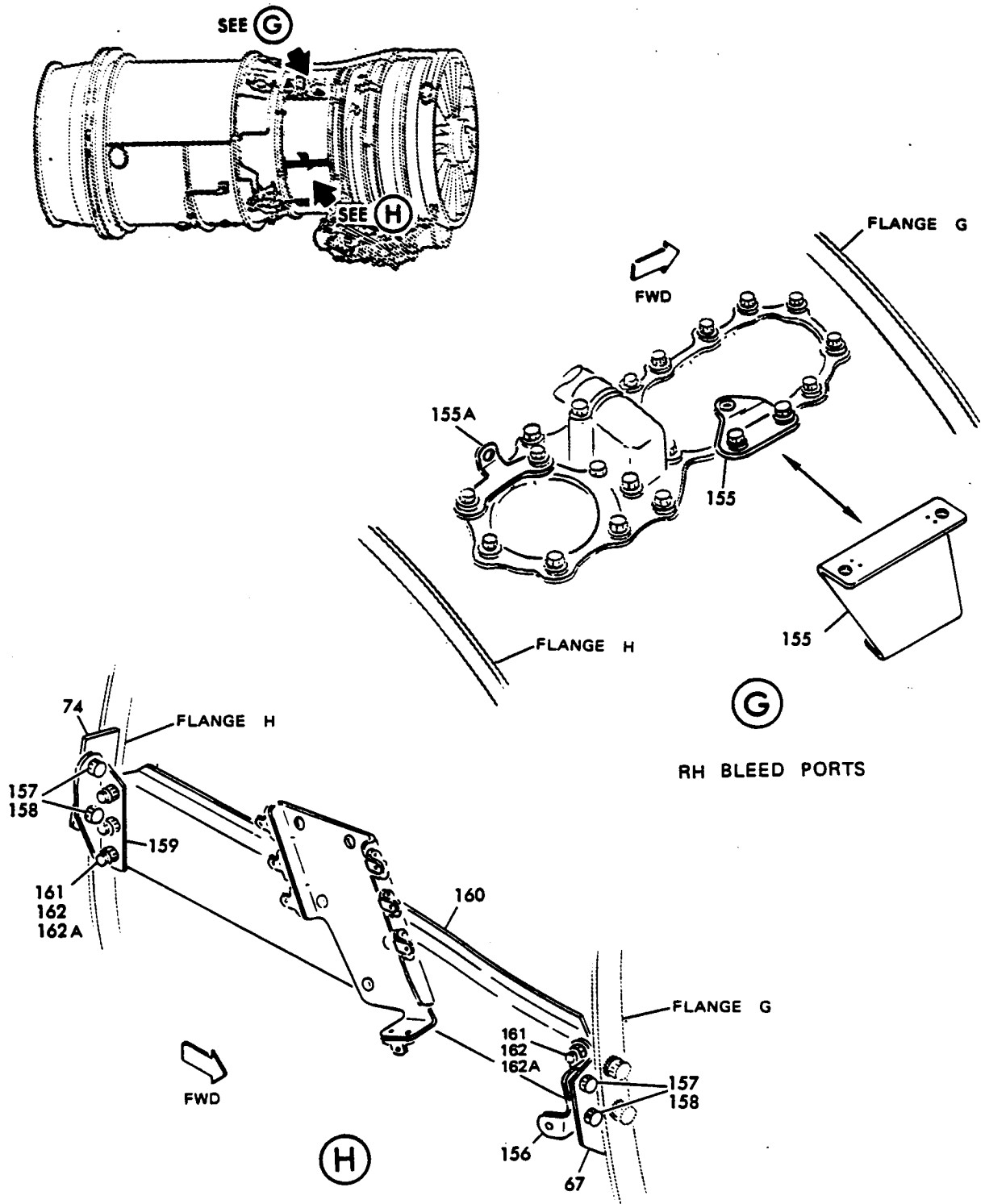
E

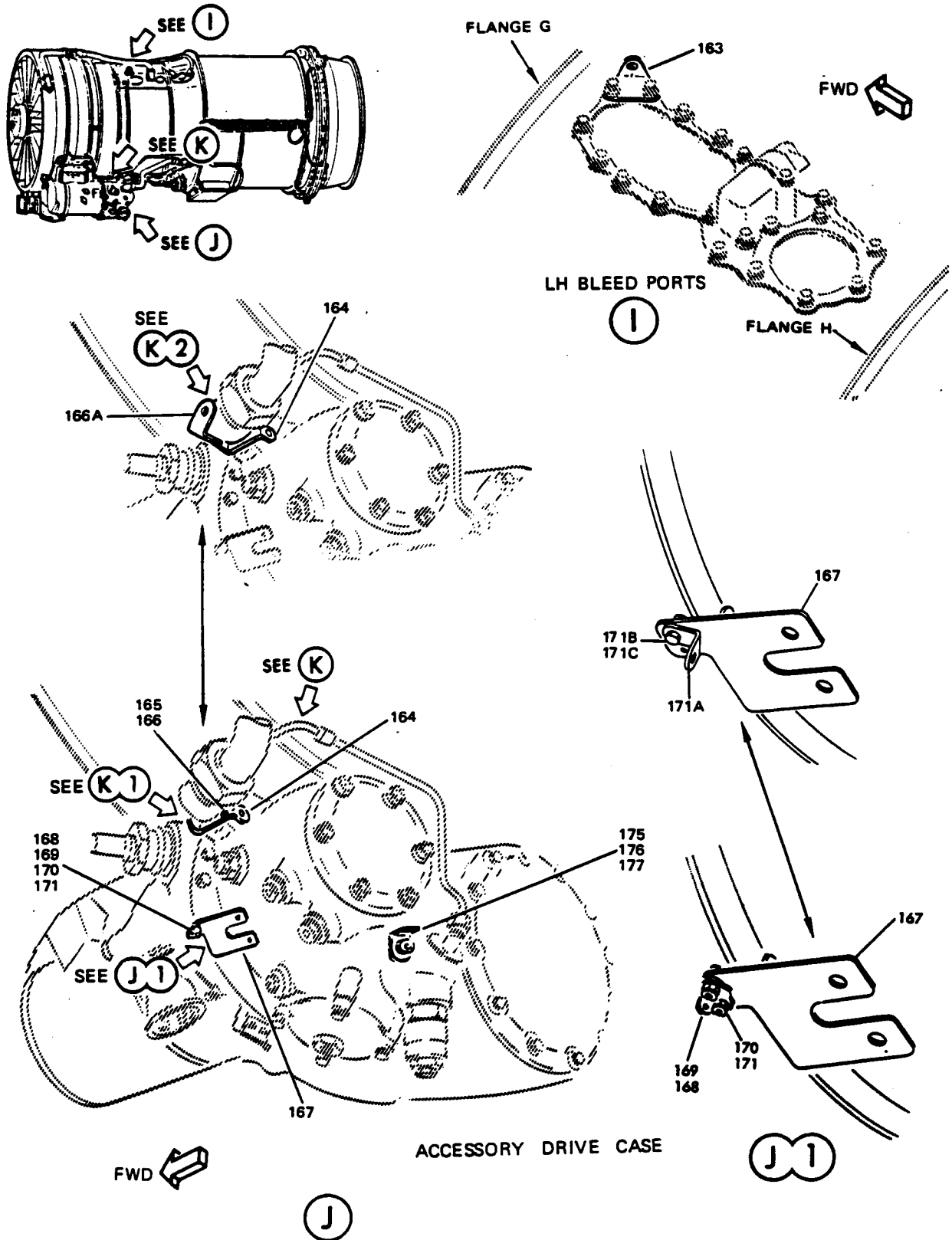
INLET ACCESSORY FLANGE



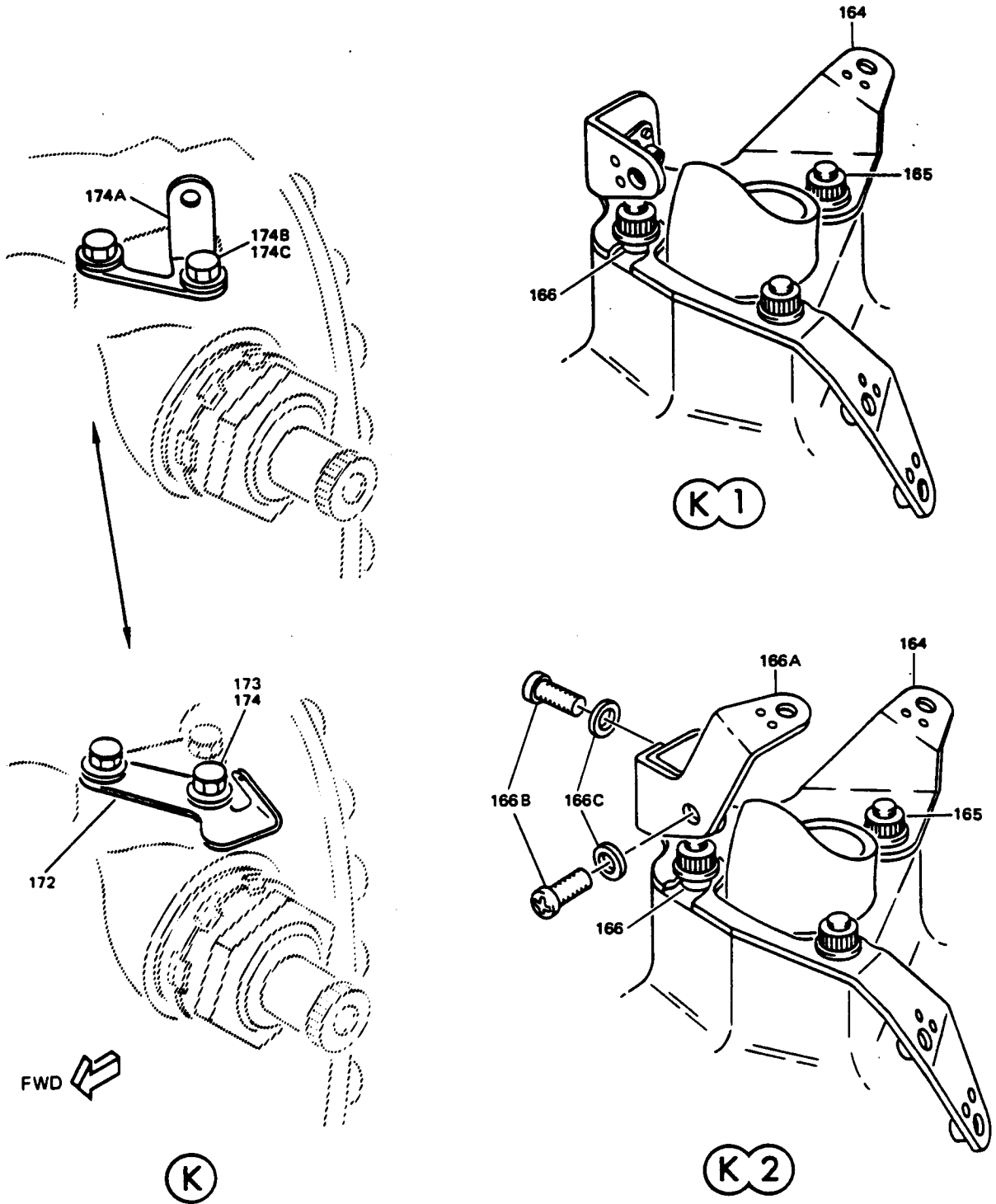
F

TAI VALVE



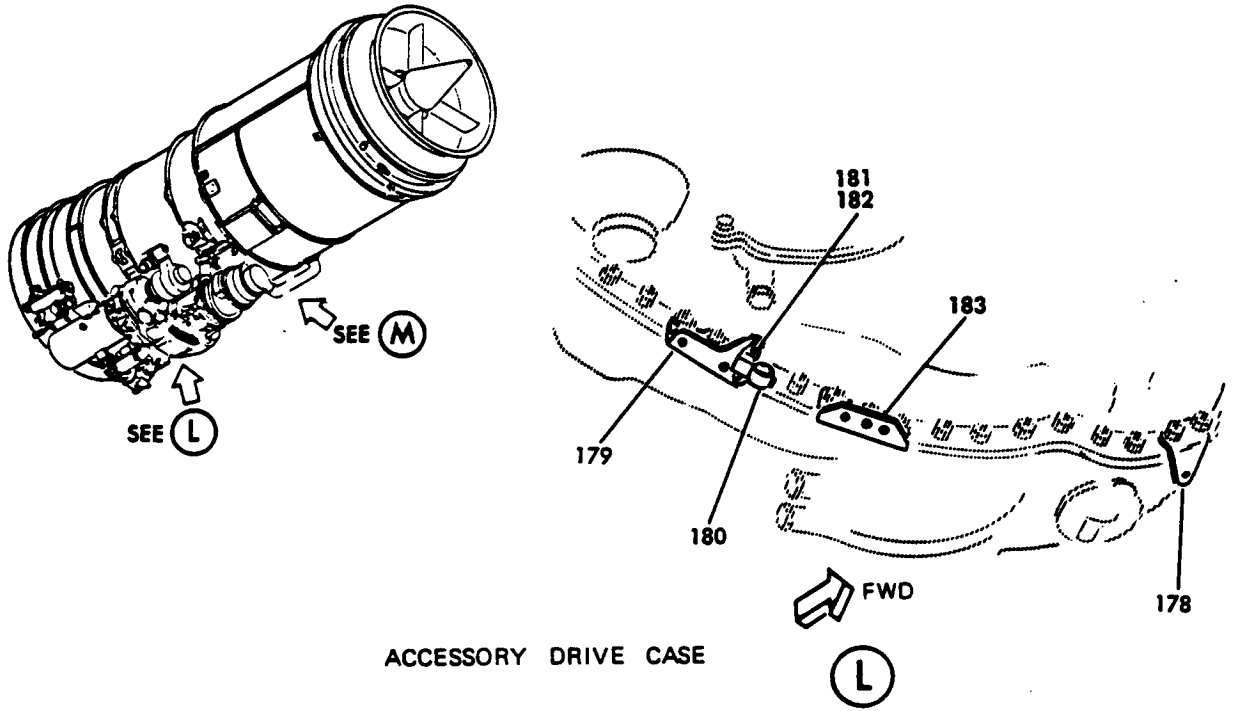


Bracket Installation
Figure 1109 (Sheet 10)

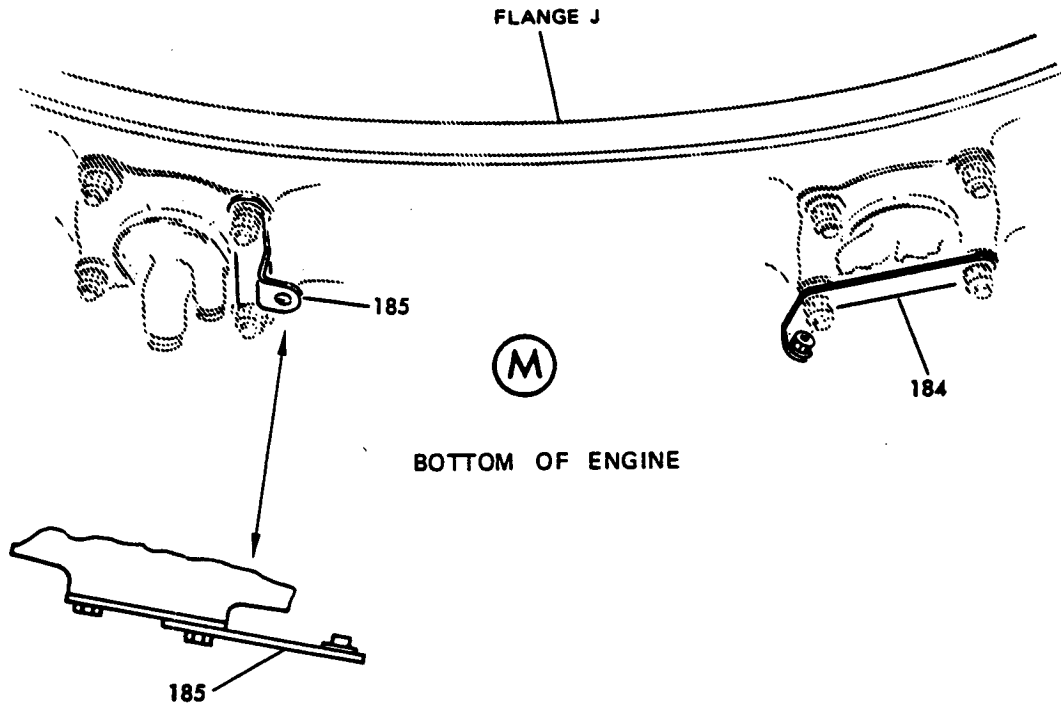


ACCESSORY DRIVE CASE

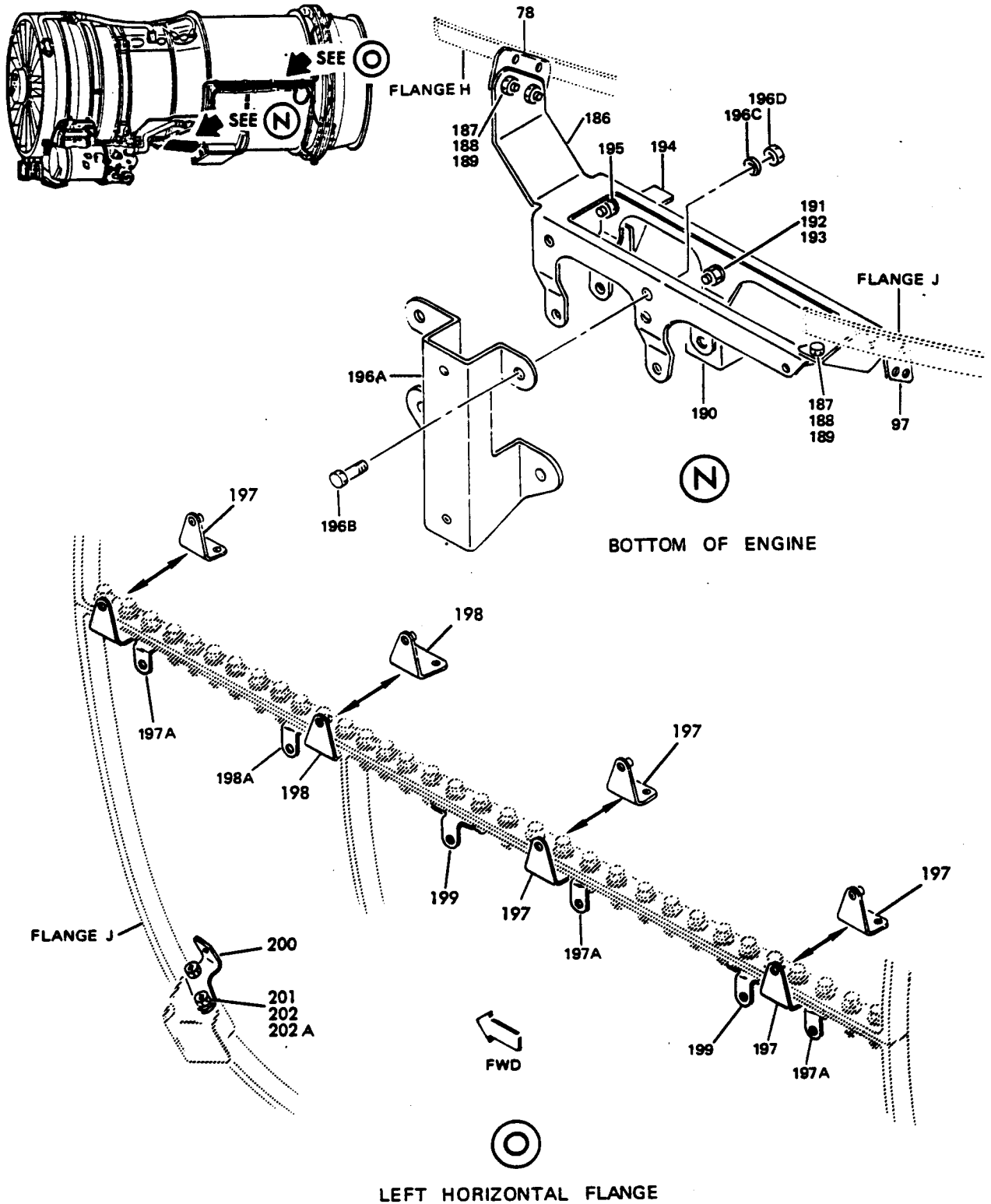
Bracket Installation
Figure 1109 (Sheet 10A)



ACCESSORY DRIVE CASE

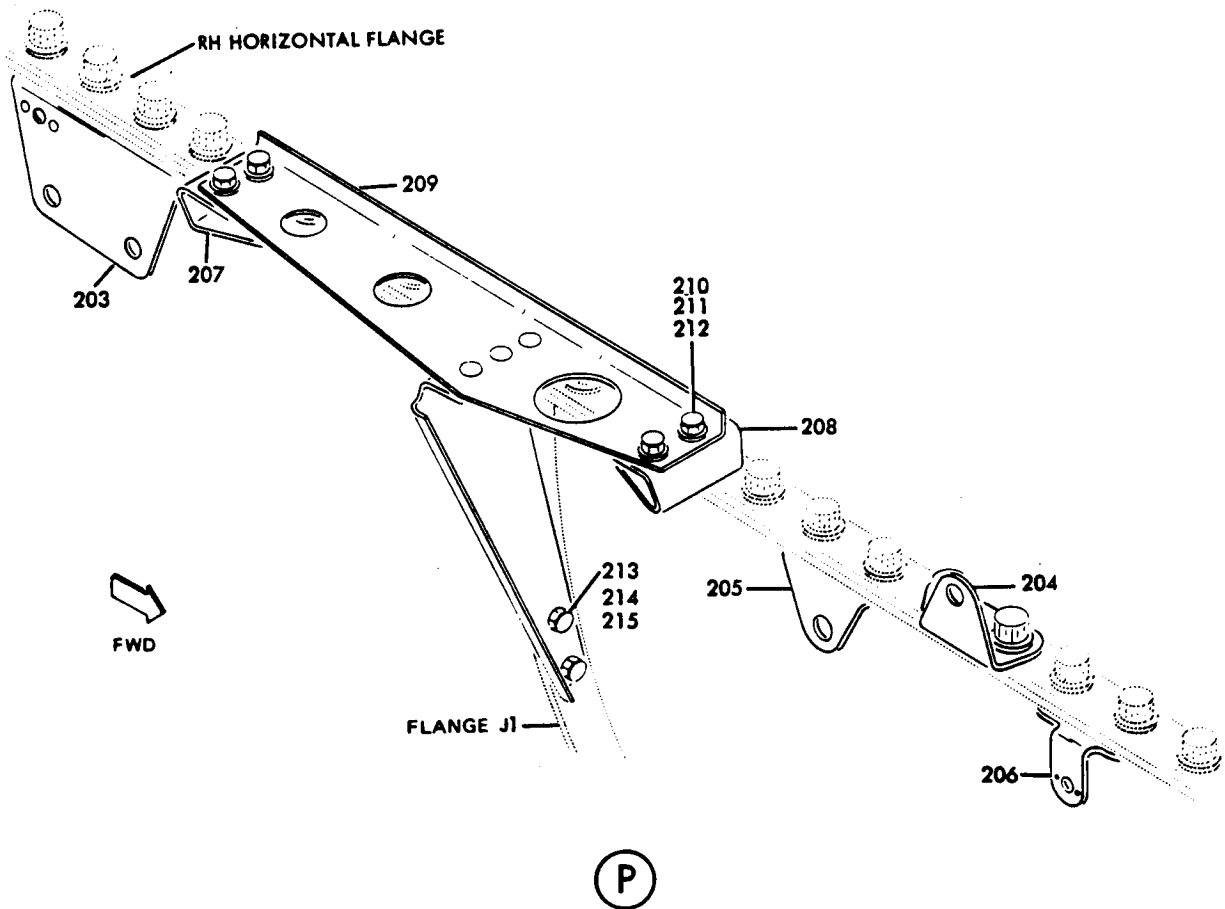
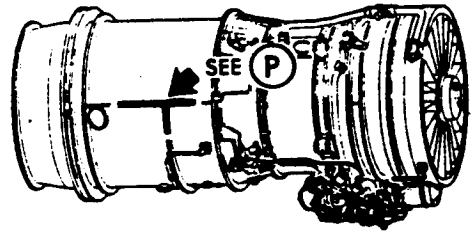


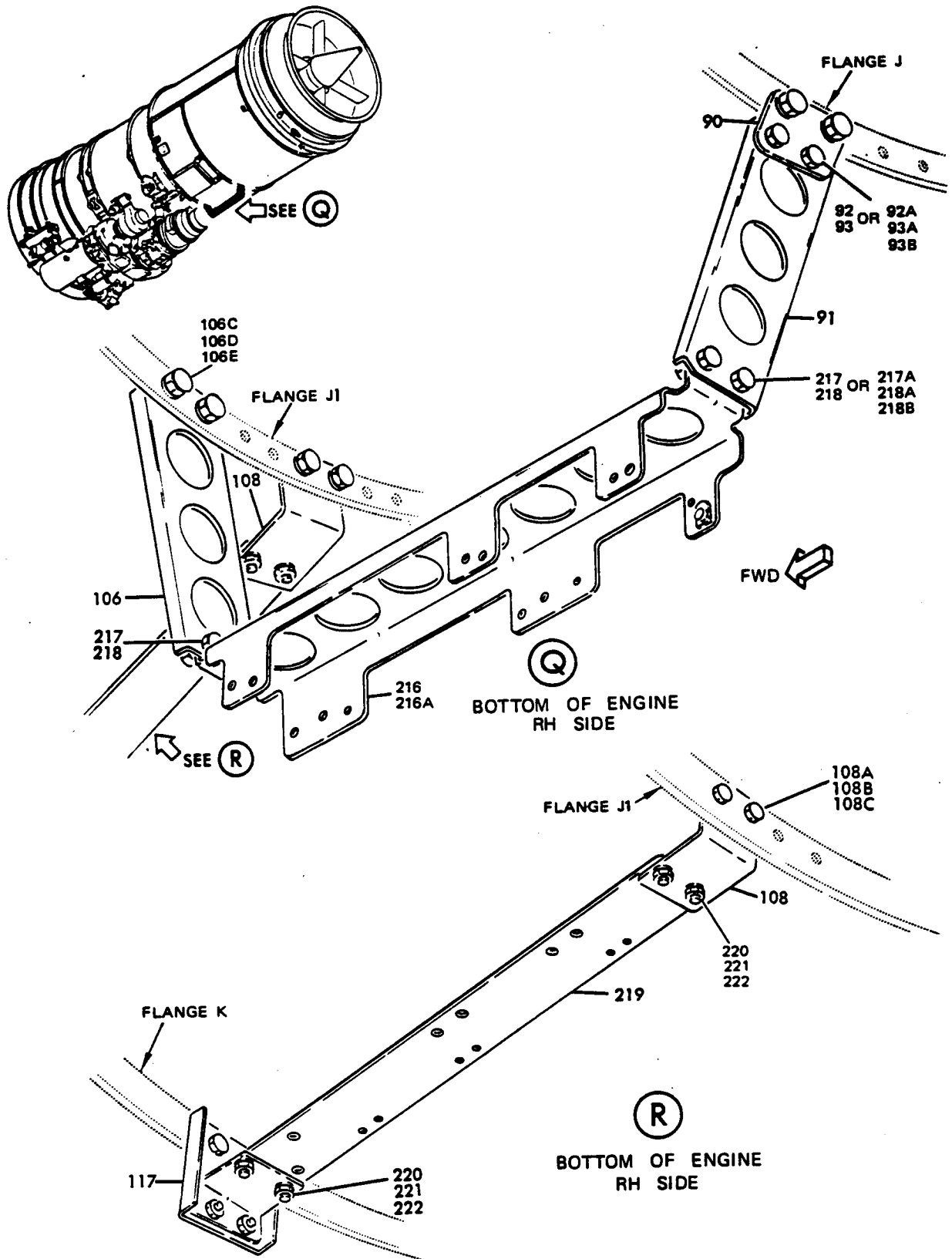
BOTTOM OF ENGINE



737
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL





Bracket Installation
Figure 1109 (Sheet 14)

737

POWERPLANT BUILDUP


BOEING
COMMERCIAL JET

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-	65-45798-1		BRACKET INSTL (BOEING) *[1]							A	
	65-45798-2		BRACKET INSTL (BOEING) *[1] *[2]							B	
	65-45798-3		BRACKET INSTL (BOEING) *[1] *[2]							C	
	65-45798-4		BRACKET INSTL (BOEING) *[1]							D	
	65-45798-5		BRACKET INSTL (BOEING) *[2] *[6]							E	
	65-45798-6		BRACKET INSTL (BOEING)							F	
	65-45798-7		BRACKET INSTL (BOEING)							G	
	65-45798-8		BRACKET INSTL (BOEING)							H	
	65-45798-9		BRACKET INSTL (BOEING) *[1]							I	
	65-45798-10		BRACKET INSTL (BOEING)							J	
	65-45799-1		BRACKET INSTL (P&W) *[1]							K	
	65-45799-117		BRACKET INSTL (P&W) *[1] *[2]							L	
	65-45799-120		BRACKET INSTL (P&W) *[1] *[2]							M	
	65-45799-123		BRACKET INSTL (P&W) *[1]							N	
	65-45799-127		BRACKET INSTL (P&W) *[1] *[2]							O	
	65-45799-134		BRACKET INSTL (P&W) *[1] *[2]							P	
	65-45799-147		BRACKET INSTL (P&W) *[1] *[2] *[6]							Q	
	65-45799-148		BRACKET INSTL (P&W) *[1] *[6]							R	
	65-45799-150		BRACKET INSTL (P&W) *[1]							S	
	65-45799-151		BRACKET INSTL (P&W) *[1]							T	
	65-45799-165		BRACKET INSTL (P&W) *[1]							U	
	65-45799-166		BRACKET INSTL (P&W) *[1]							V	
	65-45799-170		BRACKET INSTL (P&W) *[1]							W	
	65-45799-171		BRACKET INSTL (P&W) *[1]							X	
			FLANGE B								
1	464459		. BRACKET, V77445 *[23]								1
1	755775		. BRACKET, V77445 *[23]								1
2	65-45799-81		. BRACKET							K-SUV	1
2	639239		. BRACKET, V77445							K-SUV	1
2	65-45799-153		. BRACKET							LMNP-X	1
2	758083		. BRACKET, V77445							LMNP-X	1
2A	65-22486-30		. BRACKET ASSY							KLM	3
										O-SUV	
2A	463974		. BRACKET ASSY, V77445							KLM	3
										O-SUV	
2A	65-22486-258		. BRACKET ASSY							LMNP-X	3
2A	755765		. BRACKET, V77445							LMNP-X	3
3	65-22486-30		. BRACKET ASSY							K-SUV	7
3	463974		. BRACKET ASSY, V77445							K-SUV	7
3	65-22486-258		. BRACKET ASSY							LMNP-X	7
3	755765		. BRACKET, V77445							LMNP-X	7
3A	65-45799-121		. BRACKET *[43]							LMO	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-											
3A	65-45799-149		.	BRACKET ASSY (SB 71-1034) *[44]						Q	1
3A	65-45799-161		.	BRACKET *[16] *[40] *[41] *[44]						S	1
3A	65-45799-172		.	BRACKET *[39] *[42] *[44]						X	1
3B	69-60758-1		.	BRACKET *[16]						BE	1
3C	69-60758-2		.	BRACKET *[39] *[42] *[43]						B	1
3C	69-60758-3		.	BRACKET ASSY *[16] *[40] *[41] *[44]						E	1
3D	BACB30NE3-3		.	BOLT							4
3E	BACN10JC3		.	NUT							4
3F	AN960-10		.	WASHER							4
4	65-45799-29		.	BRACKET						K-SUV	1
4	581886		.	BRACKET						K-SUV	1
4	65-45799-155		.	BRACKET						LMNP-X	1
4	758085		.	BRACKET						LMNP-X	1
4A	AN960D416		.	BRACKET						K-X	2
4B	544260		.	BRACKET							1
4B	755788		.	BRACKET							1
4C	65-45799-126		.	BRACKET (OPT ON 65-45799-117, -120 PER SB 71-1015) (SB 71-1102)						OQ	1
4C	65-45799-164		.	BRACKET *[16]						SX	1
5	458075		.	BRACKET, V77445 *[23]							1
5	491633		.	BRACKET, V77445 *[23]							1
6	755782		.	BRACKET, V77445 *[23]							1
6A	65-22486-32		.	BRACKET						K-SUV	1
6A	463983		.	BRACKET, V77445						K-SUV	1
6A	65-22486-260		.	BRACKET						LMNP-X	1
6A	755771		.	BRACKET, V77445						LMNP-X	1
6B	464464		.	BRACKET, V77445 *[23]							1
6B	755788		.	BRACKET, V77445 *[23]							1
			FLANGE C								
7	65-45799-140		.	BRACKET ASSY *[00] (REPLS 65-45799-23) (SB 77-1001)						K-SUV	1
7	65-45799-23		.	BRACKET ASSY *[00] (REPLD BY 65-45799-140) (SB 77-1001)						K-SUV	1
7	583198		.	BRACKET, V77445						K-SUV	1
7	716039		.	BRACKET, V77445						K-SUV	1
7	65-45799-157		.	BRACKET						LMNP-X	1
7	758086		.	BRACKET, V77445						LMNP-X	1
8	65-22486-195		.	BRACKET ASSY						K-SUV	7

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1109-8	501776		.	BRACKET	ASSY,	V77445					K-SUV	7
8A	65-45799-152		.	BRACKET							LMNP-X	4
8B	65-45799-163		.	BRACKET							LMNP-X	1
8C	65-45799-158		.	BRACKET							LMNP-X	2
8D	698601		.	BRACKET,	V77445	*[23]						1
8E	698464		.	BRACKET,	V77445	*[23]						1
9	65-45799-43		.	BRACKET	ASSY						K-X	2
10	65-45799-805		.	BRACKET							K-RUV	1
10	65-45799-159		.	BRACKET							LMNP-X	1
10	758089		.	BRACKET,	V77445						LMNP-X	1
11	69-48205-3		.	BRACKET	ASSY						A-J	1
12	BACSL2CB3-9		.	SCREW							A-J	3
13	AN960-10L		.	WASHER							A-J	3
14	65-45799-41		.	BRACKET							K-SUV	2
14	65-45799-154		.	BRACKET							LMNP-X	2
14	758090		.	BRACKET,	V77445						LMNP-X	2
14A	65-45799-128		.	BRACKET,	VOREX DISSIPATOR						O	1
					(SB 71-1018) (OPT)							
14A	65-45799-145		.	BRACKET,	VOREX DISSIPATOR						OQ	1
					(POST SB 71-1102)							
14A	65-45799-162		.	BRACKET,	VOREX DISSIPATOR						SX	1
					(POST SB 71-1111)							
15	65-45799-103		.	BRACKET	ASSY						K-SUV	1
15	65-45799-59		.	BRACKET	ASSY (OPT)						K-SUV	1
15	583304		.	BRACKET,	V77445						K-SUV	1
15	65-45799-160		.	BRACKET							LMNP-X	1
15	758093		.	BRACKET,	V77445						LMNP-X	1
16	65-45799-48		.	BRACKET	ASSY						K-X	1
16	583306		.	BRACKET,	V77445						K-X	1
17	65-45799-16		.	BRACKET	ASSY						K-SUV	1
17	578270		.	BRACKET,	V77445						K-SUV	1
17	65-45799-156		.	BRACKET							LMNP-X	1
17	758097		.	BRACKET,	V77445						LMNP-X	1
			FLANGE D									
18	65-22486-137		.	BRACKET	ASSY	*[00]					K-SUV	2
18	480389		.	BRACKET	ASSY,	V77445					K-SUV	2
19	65-22486-206		.	BRACKET	ASSY	*[00]					K-X	3
19	491585		.	BRACKET	ASSY,	V77445					K-X	3
20	65-22486-206		.	BRACKET	ASSY	*[00]					K-X	1
20	491585		.	BRACKET	ASSY,	V77445					K-X	1
20	65-45799-65		.	DELETED								
20A	65-22486-206		.	BRACKET	ASSY						L	1
20A	491585		.	BRACKET	ASSY,	V77445					L	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY		
			1	2	3	4	5	6	7				
1109-20B	65-45799-51		.	B	R	A	C	A	S	S	KM-S UV	1	
20C	65-45799-65		.	B	R	A	C	A	S	S	K-SUV	1	
20C	647557		.	B	R	A	C	A	S	S	K-SUV	1	
20D	65-45799-65		.	B	R	A	C	A	S	S	M	1	
20D	647557		.	B	R	A	C	A	S	S	M	1	
20D	65-45799-51		.	B	R	A	C	A	S	S	L	1	
20D	583840		.	B	R	A	C	A	S	S	L	1	
20E	65-45799-65		.	B	R	A	C	A	S	S	KLN-X	1	
20E	647557		.	B	R	A	C	A	S	S	KLN-X	1	
21	65-45799-51		DELETED										
22	65-22486-206		DELETED										
22	491585		DELETED										
23	65-22486-206		.	B	R	A	C	A	S	S	K-X	3	
23	491585		.	B	R	A	C	A	S	S	K-X	3	
23A	65-22486-206		.	B	R	A	C	A	S	S	K-X	1	
23A	491585		.	B	R	A	C	A	S	S	K-X	1	
23B	65-45799-141		.	B	R	A	C	A	S	S	K-X	1	
23B	65-45799-129		.	B	R	A	C	A	S	S	K-SUV	1	
23B	709130		.	B	R	A	C	A	S	S	K-SUV	1	
24	65-45799-55		.	B	R	A	C	A	S	S	K-X	1	
24	583845		.	B	R	A	C	A	S	S	K-X	1	
24A	65-45799-130		.	B	R	A	C	A	S	S	K-X	1	
24A	709129		.	B	R	A	C	A	S	S	K-X	1	
25	65-22486-93		.	B	R	A	C	A	S	S	K-QSU -X	1	
25A	65-22486-93		.	B	R	A	C	A	S	S	RT	1	
25	463663		.	B	R	A	C	A	S	S	K-X	1	
26	65-45799-25		.	B	R	A	C	A	S	S	K-X	1	
26	583842		.	B	R	A	C	A	S	S	K-X	1	
26B	69-45402-11		.	B	R	A	C	A	S	S	U	1	
27	69-42852-1		.	B	R	A	C	A	S	S	K-X	1	
27	69-42852-800		.	B	R	A	C	A	S	S	K-X	1	
27	578366		.	B	R	A	C	A	S	S	K-X	1	
27A	NAS1104-6		.	B	O	L	T	(O	P	T	K-X	3
27B	NAS679A4W		.	N	U	T	(O	P	T	K-X	3	
27C	69-72085-2		.	B	R	A	C	A	S	S	I	1	
28	65-22486-142		.	B	R	A	C	A	S	S	K-SUV	1	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-											
35A	69-71522-1		.							U	1
35B	69-45402-12		.							U	1
35C	BACB30NF4-14		.							U	2
35D	BACN10JC4		.							U	3
35E	AN960-416L		.							U	2
35F	BACB30NF4-8		.							U	1
35G	AN960-416		.							U	2
-35J	MS20615-4M11		.	.						U	1
-35L	NAS1056C4-024		.	.						U	1
-35N	MS14101-4		.	.						U	1
-35P	69-71522-2		.	.						U	1
36	65-45799-46		.							K-X	1
36	581889		.							K-X	1
37	65-22486-114		.							K-X	1
37	467489		.							K-X	1
38	594498		.							K-X	1
38	647567		.							K-X	1
39	NAS1104-6		.							K-X	2
40	ACR10G41		.							K-X	2
41	MS20426D3-5		.							K-X	4
42	65-45799-30		.							K-X	3
42	581892		.							K-X	1
42A	65-45799-110		.							K-X	1
41A	613781		.							K-X	1
42B	BACR10G32		.							K-X	1
42C	NAS1103-3		.							K-X	1
42D	MS20426D3		.							K-X	2
43	65-45799-33		.							K-X	1
43	594497		.							K-X	1
44	NAS1104-18		.							K-X	2
45	BACR10G41		.							K-X	2
46	MS20426D3-5		.							K-X	4
47	65-45799-97		.							K-X	1
47	65-22486-87		.							K-SUV	1
47	639245		.							K-SUV	1
47	466471		.							K-SUV	1
48	65-45799-98		.							K-X	1
48	65-22486-88		.							K-SUV	1
48	466472		.							K-SUV	1
48	639247		.							K-SUV	1
49	65-45799-65		.							N	2
49	647557		.							N	2

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-50 THRU 53			. NOT USED								
			FLANGE F								
54	709140		. BRACKET ASSY V77445 (BOEING 65-45799-94)							K-X	1
54	647569		. BRACKET ASSY V77445 (BOEING 65-45799-76)(OPT)							K-SU V	1
55	NAS1306-2		DELETED								
55	BACB3ONE6-2		. BOLT							A-J	2
56	BACR10G63		. RETAINER *[00]							A-J	2
57	NAS514P440-5		DELETED								
57	NAS514P440-4		. SCREW							A-J	4
58	NAS679A04W		. NUT							A-J	4
58A	NAS1067A3		. NUTPLATE							A-J	1
58B	MS20426D3-4		. RIVET							A-J	2
59	65-22486-95		. BRACKET							K-X	2
59	466630		. BRACKET, V77445							K-X	2
60	65-22486-148		. BRACKET							K-X	1
60	490479		. BRACKET, V77445							K-X	1
61	65-45799-22		. BRACKET ASSY							K-X	1
61	584288		. BRACKET, V77445							K-X	1
62	65-22486-230		. BRACKET							K-X	1
62	567326		. BRACKET, V77445							K-X	1
63	65-45799-39		. BRACKET ASSY							K-X	1
			FLANGE G								
64	65-45799-52		. BRACKET *[15]*[24]*[25]							KNP-X	1
64	583477		. BRACKET, V77445							KNP-X	1
64	65-45799-118		. BRACKET							LM	1
64	65-45799-132		. BRACKET (OPT)							O	
64	65-45799-144		. BRACKET *[16]							OQ	1
64	65-45799-174		. BRACKET *[26]							QSWX	1
65	65-22486-173		. BRACKET ASSY							K-X	1
65	491578		. BRACKET ASSY, V77445							K-X	1
66	65-45799-24		. BRACKET ASSY							K-X	2
66	639256		. BRACKET, V77445							K-X	1
66A	65-45799-241		DELETED								
67	65-45799-28		. BRACKET ASSY							K-X	1
67	577994		. BRACKET, V77445							K-X	1
68	65-45799-34		. BRACKET (PRE SB 36-1013)							K-X	1
68	583476		. BRACKET, V77445							K-X	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-68	65-45799-176		.	B	R	A	C	K	E	X	1
68A	69-72085-2		.	B	R	A	C	K	E	I	1
68B	69-72085-1		.	B	R	A	C	K	E	I	1
68C	69-66635-1		.	B	R	A	C	K	E	D	1
68D	BACS12CB3-7		.	S	C	R	E	W		D	2
68E	AN960D10L		.	W	A	S	H	E		D	2
68F	69-39688-1		.	B	R	A	C	K	E	A-J	1
68G	69-22313-12		.	B	R	A	C	K	E	A-J	1
68H	65-45799-173		.	B	R	A	C	K	E	K-X	1
68H	65-45799-175		.	B	R	A	C	K	E	K-X	1
			F L A N G E H								
69	65-45799-36		.	B	R	A	C	K	E	K-X	1
70	65-45799-74		.	B	R	A	C	K	E	K-SUV	1
70	639260		.	B	R	A	C	K	E	K-SUV	1
71	65-45799-6		.	B	R	A	C	K	E	K-X	1
71A	65-45799-8		.	.	B	R	A	C	K		1
72	65-45799-9		.	.	B	R	A	C	K		1
72A	NAS1104-12		.	.	B	O	L	T	(O	P	4
72B	NAS679A4W		.	.	N	U	T	(O	P		4
73	65-45799-82		.	B	R	A	C	K	E	K-X	1
73	639258		.	B	R	A	C	K	E	K-X	1
74	65-22486-50		.	B	R	A	C	K	E	K-X	1
74	463717		.	B	R	A	C	K	E	K-X	1
75	65-45799-40		.	B	R	A	C	K	E	K-X	1
75	581736		.	B	R	A	C	K	E	K-X	1
76	65-45799-50		.	B	R	A	C	K	E	K-SUV	1
76	581738		.	B	R	A	C	K	E	K-SUV	1
77	65-45799-37		.	B	R	A	C	K	E	K-X	
77	581739		.	B	R	A	C	K	E	K-X	1
78	65-22486-8		.	B	R	A	C	K	E	K-X	1
78	463712		.	B	R	A	C	K	E	K-X	1
78A	65-45799-142		.	B	R	A	C	K	E	K-X	1
78A	716038		.	B	R	A	C	K	E	K-X	1
79	69-43416-1		.	F	I	T	T	I	N	K-X	2
79	581734		.	B	R	A	C	K	E	K-X	2
80	65-45799-5		.	B	R	A	C	K	E	K-X	1
80A	65-45799-7		.	.	B	R	A	C	K		1
81	65-45799-9		.	.	B	R	A	C	K		1
81A	NAS1104-12		.	.	B	O	L	T	(O	P	4
81B	NAS679A4W		.	.	N	U	T	(O	P		4

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY		
			1	2	3	4	5	6	7				
1109-													
82	65-45799-64										. BRACKET ASSY	K-X	1
82	581733										. BRACKET, V77445	K-X	1
82A	69-33197-2										. BRACKET ASSY (POST SB 79-1004)	J	1
82B	NAS1801-3-7										. SCREW (POST SB 79-1004)	J	3
82C	AN960D10L										. WASHER (POST SB 79-1004)	J	3
82D	BACN10JC3										. NUT (POST SB 79-1004)	J	3
82E	65-22486-135										. BRACKET (POST SB 79-1004)	K-X	1
82E	491631										. BRACKET, V77445	K-X	1
82F	69-22313-13										. BRACKET	A-J	1
82G	69-39688-1										. BRACKET	A-J	1
82H	69-58497-6										. BRACKET	A-J	1
82I	69-58497-1										. BRACKET	A-J	1
82J	69-20607-6										. BRACKET	A-J	1
82K	69-24228-1										. BRACKET	IJ	1
82L	69-45428-3										. BRACKET	A-J	1
											FLANGE J		
83	65-45799-27										. BRACKET ASSY	K-X	5
83	451094										. BRACKET, V77445	K-X	5
84	65-45799-72										. BRACKET ASSY	K-X	1
84	639261										. BRACKET, V77445	K-X	1
85	65-45799-21										. BRACKET ASSY	K-X	1
85	584296										. BRACKET, V77445	K-X	1
86	65-45799-27										. BRACKET ASSY	K-X	1
86	451094										. BRACKET, V77445	K-X	1
87	65-45799-99										. BRACKET	K-X	1
87	639264										. BRACKET, V77445	K-X	1
87	65-45799-81										. BRACKET	K-X	1
87	639239										. BRACKET, V77445	K-X	1
88	69-41780-1										. BRACKET	K-X	1
88A	NAS1104-9										. BOLT (OPT)	K-X	2
88B	NAS679A4W										. NUT (OPT)	K-X	2
89	65-45799-47										. BRACKET ASSY	K-X	1
89	584289										. BRACKET, V77445	K-X	1
90	65-45799-100										. BRACKET	K-X	1
90	639263										. BRACKET, V77445	K-X	1
91	69-52295-2										. BRACKET (RELOCATE FOR 65-45798-7 ONLY)	G	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-											
92	BACS12CB-3-6		.	S	C	R	E	W		K-X	2
92A	NAS1801-3-14		.	S	C	R	E	W		G	2
93	NAS679A3W		.	N	U	T				K-X	2
93A	BACN10JC3		.	N	U	T				G	2
93B	NAS43DD3-32		.	S	P	A	C	E	R	G	2
94	65-45799-47		.	B	R	A	C	K	E	T	1
94	584289		.	B	R	A	C	K	E	T	1
95	65-45799-3		.	B	R	A	C	K	E	T	1
95	584294		.	B	R	A	C	K	E	T	1
96	69-41777-800		.	B	R	A	C	K	E	T	1
96	578837		.	B	R	A	C	K	E	T	1
96	69-41777-1		.	B	R	A	C	K	E	T	1
96A	NAS1304-9		.	B	O	L	T			K-X	1
97	65-22486-16		.	B	R	A	C	K	E	T	1
97	466782		.	B	R	A	C	K	E	T	1
97A	69-20607-6		.	B	R	A	C	K	E	T	1
97A	69-58497-1		.	B	R	A	C	K	E	T	1
97A	69-58497-6		.	B	R	A	C	K	E	T	1
98	65-22486-33		.	B	R	A	C	K	E	T	1
98	463749		.	B	R	A	C	K	E	T	1
99	69-45428-10		.	B	R	A	C	K	E	T	1
99	69-45428-1		.	B	R	A	C	K	E	T	1
99	69-45428-12		.	B	R	A	C	K	E	T	1
100	BACS12CB3-6		.	S	C	R	E	W		A-J	2
101	NAS679A3W		.	N	U	T				A-J	2
102	65-22486-33		.	B	R	A	C	K	E	T	2
102	463749		.	B	R	A	C	K	E	T	2
102A	69-45402-2		.	B	R	A	C	K	E	T	1
102B	BACS12CB3-6		.	S	C	R	E	W		A-J	2
102C	NAS679A3W		.	N	U	T				A-J	2
102D	AN960D10L		.	W	A	S	H	E	R	A-J	2
103	65-22486-33		.	B	R	A	C	K	E	T	2
103	463749		.	B	R	A	C	K	E	T	2
103A	AN960-416		.	W	A	S	H	E	R	K-X	1
104	65-45799-15		.	B	R	A	C	K	E	T	1
104	584298		.	B	R	A	C	K	E	T	1
104	65-45799-109		.	B	R	A	C	K	E	T	1
			.	B	R	A	C	K	E	T	1
104	613780		.	B	R	A	C	K	E	T	1
104A	540966		.	B	R	A	C	K	E	T	1

737

POWERPLANT BUILDUP



FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-			FLANGE J1								
105	69-45446-4		. BRACKET ASSY							A-J	1
105A	NAS1104-3		. BOLT							A-J	2
105B	NAS679A4W		. NUT							A-J	2
105C	AN960D416		. WASHER							A-J	2
106	69-52295-1		. BRACKET							A-J	1
106A	BACN10MK3-85		. NUTPLATE							A-J	1
106B	MS20615-4M		. RIVET							A-J	2
106C	NAS1104-3		. BOLT							A-J	2
106D	AN960D416		. WASHER							A-J	2
106E	NAS679A4W		. NUT							A-J	2
107	69-45492-3		. BRACKET ASSY							A-J	1
107A	NAS1104-3		. BOLT							A-J	2
107B	AN960D416		. WASHER							A-J	2
107C	NAS679A4W		. NUT							A-J	2
108	69-52283-3		. BRACKET ASSY *[13]*[17]*[29]*[33] *[34]*[36]							A	1
108	69-62249-1		. BRACKET ASSY *[14]*[18]*[30]*[35] *[37]							B-J	1
108A	NAS1104-3		. BOLT							A-J	2
108B	AN960D416		. WASHER							A-J	2
108C	NAS69A4W		. NUT							A-J	2
109	69-45492-2		. BRACKET ASSY							A-J	1
109A	NAS1104-3		. BOLT							A-J	3
109B	AN960D416		. WASHER							A-J	3
109C	NAS679A4W		. NUT							A-J	3
110	69-45428-14		. BRACKET							A-J	1
110	69-45428-4		. BRACKET (OPT)							A-J	1
110A	NAS1104-3		. BOLT							A-J	2
110B	AN960D416		. WASHER							A-J	2
110C	NAS679A4W		. NUT							A-J	2
111	69-43417-1		. BRACKET							A-J	1
111A	NAS1104-4		. BOLT							A-J	2
111B	AN960D416		. WASHER							A-J	2
111C	NAS679A4W		. NUT							A-J	2

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-			FLANGE K								
112	65-45799-90		. BRACKET							K-X	1
112	639269		. BRACKET, V77445							K-X	1
113	65-45799-800		. BRACKET ASSY *[00]							K-SUV	6
113	454423		. BRACKET ASSY, V77445							K-SUV	6
114	65-46799-11		. BRACKET ASSY *[13]*[29]*[36]							K-O	1
										Q-SUV	
114	65-45799-137		. BRACKET ASSY *[14]*[30]*[37]							K-X	1
114A	65-45799-13		. . BRACKET (POST SB 71-1032) [14]							K-O	1
										Q-SUV	
114B	65-45799-12		. . BRACKET							K-O	1
										Q-SUV	
114B	65-45799-135		. . BRACKET *[14]							K-X	1
114C	65-45799-11		. BRACKET ASSY *[13]*[29]*[36]							K-O	1
										Q-SUV	
114C	65-45799-138		. BRACKET ASSY *[14]*[30]*[37]							K-X	1
114D	65-45799-13		. . BRACKET *[14]							K-O	1
										Q-SUV	
114E	65-45799-12		. . BRACKET							K-O	1
										Q-SUV	
114E	65-45799-136		. . BRACKET *[14]							K-X	1
115	NAS1104-16		. BOLT (OPT WITH 65-45799-11)							K-O	6
										Q-SUV	
115	NAS1104-13		. BOLT (OPT WITH 65-45799-137 AND 65-45799-138)							K-X	6
115	BACB30NF4-13		. BOLT *[14]							K-X	6
115A	NAS679A4W		. NUT *[14]							K-X	6
116	65-45799-71		. BRACKET *[00]							K-SUV	1
116	639268		. BRACKET, V77445							K-SUV	1
117	65-45799-105		. BRACKET *[00]							K-X	1
117	639267		. BRACKET, V77445							K-X	1
117	65-45799-19		. BRACKET *[00]							K-SUV	1
117	578494		. BRACKET, V77445							K-SUV	1
117A	69-66601-1		. BRACKET (SB 26-1014)							K-SUV	1
118	65-45799-49		. BRACKET ASSY							K-X	1
118	584219		. BRACKET, V77445							K-X	1
119	65-45799-77		. BRACKET ASSY *[3]							K-X	1
119	639266		. BRACKET, V77445							K-X	1
120	65-45799-17		. BRACKET ASSY							K-X	1
120	584216		. BRACKET, V77445							K-X	1
121	65-45799-800		. BRACKET ASSY *[00]							K-SUV	1
121	454423		. BRACKET ASSY, V77445							K-SUV	1
122	65-45799-800		. BRACKET ASSY *[00]							K-SUV	5
122	454423		. BRACKET ASSY, V77445							K-SUV	5

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1109-												
123	65-22486-86		.	BRACKET ASSY	*[00]					K-SUV	1	
123	463649		.	BRACKET ASSY	*[00]					K-SUV	1	
123A	65-45799-800		.	BRACKET ASSY	*[00]					K-SUV	1	
123A	454423		.	BRACKET ASSY	*[00]					K-SUV	1	
124	65-22486-35		.	BRACKET ASSY	*[00]					K-SUV	1	
124	463647		.	BRACKET ASSY	*[00]					K-SUV	1	
124A	65-45799-800		.	BRACKET ASSY	*[00]					K-SUV	2	
124A	454423		.	BRACKET ASSY	*[00]					K-SUV	2	
				REAR ENGINE MOUNT FLANGE								
125	69-43468-1		.	BRACKET						A-J	2	
126	69-43487-1		.	BUSHING						A-J	8	
127	NAS1104-5		.	BOLT						A-J	8	
128	NAS679A4W		.	NUT						A-J	8	
129	AN960C416L		.	WASHER						A-J	AR	
130	AN960C416		.	WASHER						A-J	8	
131	69-29863-2		.	RUB STRIP						A-J	1	
132	NAS1105-3		.	BOLT						A-J	1	
133	BACW10AK-WP5		.	WASHER						A-J	1	
134	BACW10P16S		.	WASHER						A-J	1	
135	NAS679A5		.	NUT						A-J	1	
136	69-48235-1			DELETED								
137	65-45798-800			DELETED								
138	65-45798-801			DELETED								
139	65-45798-802			DELETED								
140	NAS1104-6			DELETED								
141	NAS679A4			DELETED								
				INLET ACCESSORY FLANGE								
142	65-22486-169		.	BRACKET ASSY						K-X	1	
142	468904		.	BRACKET ASSY, V77445						K-X	1	
				TAI VALVE								
143	65-45799-38		.	BRACKET ASSY						K-X	1	
143	584327		.	BRACKET, V77445						K-X	1	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-			FLANGE D & E - LH								
144	69-25331-8		. BRACKET *[00]							A-HJ	1
144	69-60439-2		. BRACKET ASSY, OIL PRESSURE TRANSMITTER SUPPORT (SB 79-1002)							A-HJ	1
145	69-25331-1		. BRACKET ASSY							A-HJ	1
146	69-25331-5		. BRACKET *[00]							A-HJ	1
146	69-60439-1		. BRACKET ASSY, OIL PRESSURE TRANSMITTER SUPPORT (SB 79-1002)							A-HJ	1
147	69-25331-2		. BRACKET ASSY							A-HJ	1
148	BACS12CB3-7		. SCREW							A-HJ	6
149	AN960D10L		. WASHER							A-HJ	6
150	NAS679A3W		. NUT							A-HJ	2
			FUEL CONTROL MOUNT FLANGE								
151	65-45799-101		. BRACKET							K-X	1
151	214335-8		. BRACKET, V59875							K-X	1
152	TA12100004		. CLAMP ASSY, V84971							K-X	1
152	TA5000BH8AT		. CLAMP ASSY, V84971							K-X	1
152	TA5000BH8A		. CLAMP ASSY, V84971							K-X	1
153	BACS12CB3-6		. SCREW							K-X	2
154	NAS679A3W		. NUT							K-X	2
			RH BLEED PORTS								
155	65-45799-57		. BRACKET ASSY *[15]							KPR T-W	1
155	584320		. BRACKET, V77445							KPR T-W	1
155	65-45799-122		. BRACKET							LM	1
155	65-45799-133		. BRACKET ASSY							O	1
155	65-45799-143		. BRACKET ASSY *[16]							OQSX	1
155A	69-48235-4		. BRACKET ASSY (USED ONLY WITH 65-73778 THRUST REVERSER INSTL)								1
			FLANGE G TO H - RH SIDE								
156	69-22313-12		. BRACKET							A-J	1
157	BACS12CB3-6		. SCREW							A-J	4
158	NAS679A3W		. NUT							A-J	4
159	69-22313-13		. BRACKET							A-J	1
160	69-39688-1		. BRACKET ASSY							A-J	1
161	BACS12CB3-7		. SCREW							A-J	5
162	NAS679A3W		. NUT							A-J	5
162A	AN960D10L		. WASHER							A-J	5

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-			LH BLEED PORTS								
163	65-45799-802		. BRACKET *[00]							K-SUV	1
163	584319		. BRACKET, V77445							K-SUV	1
			BRACKETS ON ACCESSORY DRIVE CASE								
164	69-45402-7		. BRACKET ASSY							A-J	1
165	NAS679A4W		. NUT							A-J	3
166	NAS43HT4-11		. SPACER							A-J	3
166A	69-45402-10		. BRACKET ASSY (POST SB 79-1004)							FJ	1
166B	BACS12CB3-7		. SCREW							FJ	2
166B	NAS1801-3-9		. SCREW (POST SB 79-1004)							FJ	2
166C	AN960-10L		. WASHER (POST SB 79-1004)							FJ	2
167	69-53880-1		. BRACKET *[00]							K-T	1
										VWX	
167	69-53885-1		. BRACKET *[00]							K-T	1
										VWX	
167	69-53885-2		. BRACKET							U	1
168	AN960-416		. WASHER							K-X	6
169	BACN10MK3-45		. NUTPLATE *[7]							A-E	1
										G-J	
170	BACS12CB06-6		. SCREW *[7]							A-E	2
										G-J	
171	NAS679A06W		. NUT *[7]							A-E	2
										G-J	
171A	AN743-BP12		. BRACKET							F	1
171B	BACS12CB3-7		. SCREW							F	1
171C	AN960PD10L		. WASHER							F	1
171D	BACN10JC3		. NUT							F	1
172	69-45804-4		. BRACKET ASSY							A-E	1
										G-J	
173	NAS1304-2H		. BOLT							A-E	2
										G-J	
174	AN960D416		. WASHER							A-E	2
										G-J	
174A	69-45402-9		. BRACKET ASSY							F	1
174B	NAS6604H2		. BOLT							F	2
174C	AN960-416		. WASHER							F	2
175	AN743P12		. ANGLE							A-J	1
176	NAS1303-1H		. BOLT							A-J	1
177	AN960D10L		. WASHER							A-J	1
178	65-45799-93		. BRACKET							K-X	1
178	647562		. BRACKET, V77445							K-X	1
179	65-45799-102		. BRACKET ASSY *[00]							K-X	1
179	647560		. BRACKET, V77445							K-X	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-											
180	TA12100004		.							K-X	1
180	TA5000BH8AT		.							K-X	1
180	TA5000BH8A		.							K-X	1
181	BACS12CB3-6		.							K-X	2
182	NAS679A3W		.							K-X	2
183	65-45799-26		.							K-X	1
183	584309		.							K-X	1
			BOTTOM OF ENGINE FWD OF J FLANGE								
184	65-45799-20		.							K-X	1
184	584325		.							K-X	1
185	65-45799-45		.							K-UWX	1
185	584323		.							K-UWX	1
185	65-45799-167		.							V	1
			BOTTOM LH BETWEEN FLANGE J AND H								
186	69-20607-6		.							A-J	1
186	69-58497-1		.							A-J	1
186	69-58497-6		.							A-J	1
187	BACS12CB3-7		.							A-J	4
188	AN960D10L		.							A-J	4
189	NAS679A3W		.							A-J	4
190	69-45428-3		.							A-J	1
190	69-65869-3		.							G	1
191	BACS12CB3-8		.							A-J	1
192	AN960D10L		.							A-J	1
193	NAS679A3W		.							A-J	1
194	69-45428-8		.							A-J	1
195	BACS12CB3-8		.							A-J	1
196			.							DELETED	
196A	69-24228-1		.							IJ	1
196B	NAS1801-3-8		.							IJ	2
196C	AN960D10L		.							IJ	2
196D	BACN10JC3		.							IJ	2
			LEFT HORIZONTAL FLANGE								
197	65-45799-124		.							K-X	3
197	511806		.							K-X	3
197A	65-45799-75		.							K-SUV	3
197A	516008		.							K-SUV	3

737

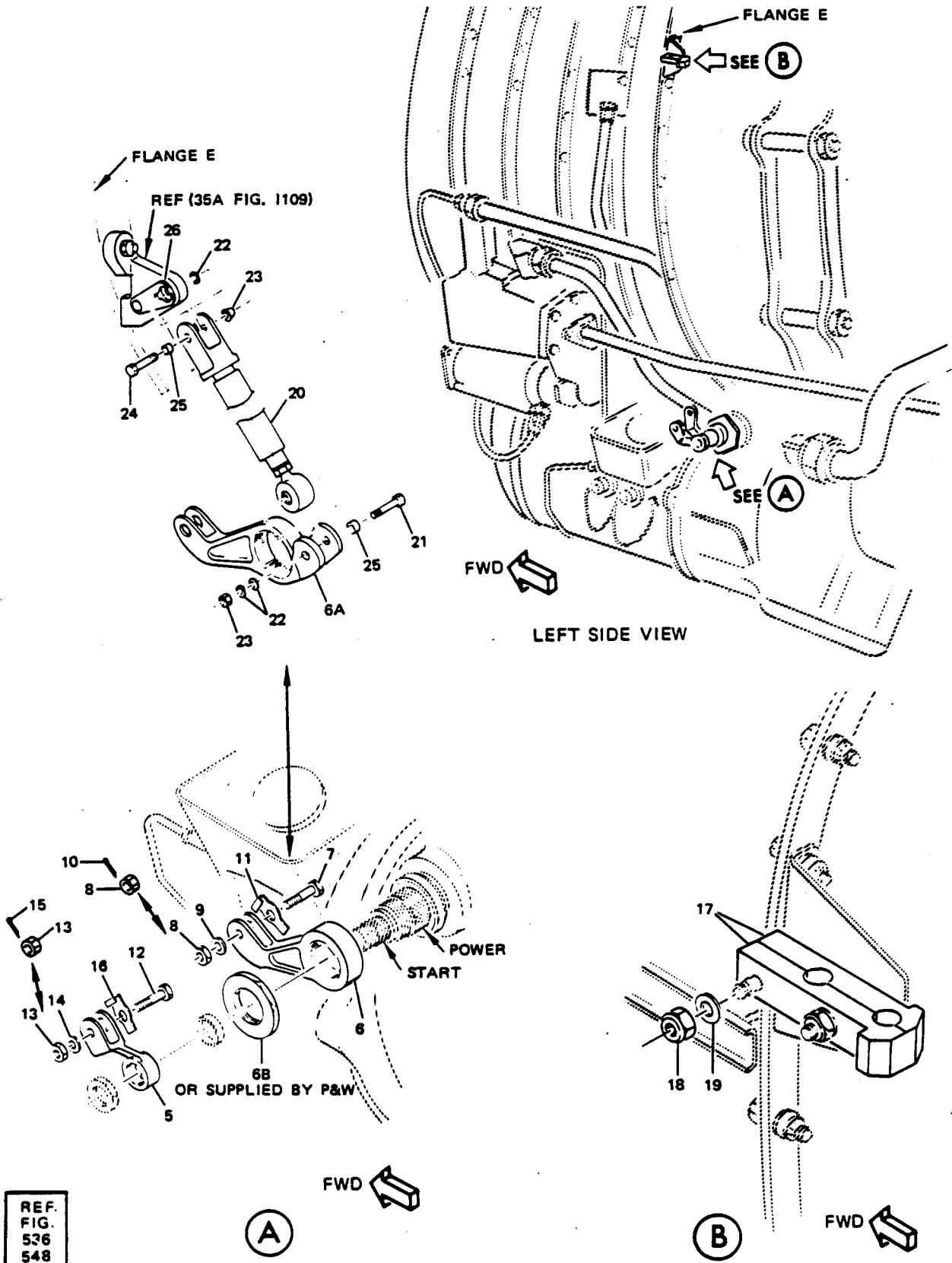
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1109-												
198	65-45799-125		.	BRACKET	ASSY	*[00]				K-X	1	
198	511806		.	BRACKET	ASSY,	V77445				K-X	1	
198A	65-45799-92		.	BRACKET	ASSY	*[00]				K-SUV	1	
198	511804		.	BRACKET	ASSY,	V77445				K-SUV	1	
199	65-45799-801		.	BRACKET	ASSY	*[00]				K-SUV	2	
199	454425		.	BRACKET	ASSY,	V77445				K-SUV	2	
200	69-45402-2		.	BRACKET	ASSY					A-J	1	
201	BACS12CB3-6		.	SCREW						A-J	2	
202	NAS679A3W		.	NUT						A-J	2	
202A	AN960D10L		.	WASHER						A-J	2	
				RIGHT HORIZONTAL FLANGE								
203	65-45799-80		.	BRACKET	ASSY					K-X	1	
203	639275		.	BRACKET,	V77445					K-X	1	
204	65-45799-875		.	BRACKET						K-SUV	1	
204	639273		.	BRACKET,	V77445					K-SUV	1	
204	65-45799-168		.	BRACKET						K-X	1	
205	65-45799-876		.	BRACKET						K-SUV	1	
205	639274		.	BRACKET,	V77445					K-SUV	1	
205	65-45799-169		.	BRACKET						K-X	1	
206	65-45799-801		.	BRACKET	ASSY	*[00]				K-X	1	
206	454425		.	BRACKET	ASSY,	V77445				K-X	1	
207	65-45799-4		.	BRACKET						K-X	1	
207	581728		.	BRACKET,	V77445					K-X	1	
208	65-45799-10		.	BRACKET						K-X	1	
208	581727		.	BRACKET,	V77445					K-X	1	
209	69-36634-4		.	BRACKET	ASSY (USED ON 65-45798-1)					A	1	
209	69-36634-4		.	BRACKET (SB 71-1058, 71-1114)*[12]						A		
				[13][19]*[22]*[28]*[34]*[36]								
209	69-62271-1		.	BRACKET	ASSY (USED ON 65-45798-3, 6,-7,-8,-9)*[10]					B-J	1	
209	69-62271-1		.	BRACKET (SB 71-1058, 71-1114)*[10]						B-J	1	
				[14][20]*[21]*[27]*[35]*[37]								
210	BACS12CB3-7		.	SCREW						A-J	4	
211	AN960D10L		.	WASHER						A-J	4	
212	NAS679A3W		.	NUT						A-J	4	
213	NAS1104-4		.	BOLT						A-J	2	
214	AN960D416		.	WASHER						A-J	2	
215	NAS679A4W		.	NUT						A-J	2	

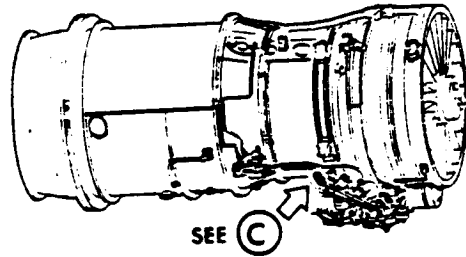
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1109-			BOTTOM RH - FLANGE J AND J1								
216	69-52261-1		. SUPPORT ASSY (LIMITED USAGE)							A-J	1
216A	69-52261-3		. SUPPORT ASSY (LIMITED USAGE)							A-J	1
217	BACS12CB3-6		. SCREW							A-FH-J	4
217A	NAS1801-3-14		. SCREW							G	2
218	NAS679A3W		. NUT							A-FH-J	4
218A	BACN10JC3		. NUT							G	2
218B	NAS43DD3-32		. SPACER							G	2
			BOTTOM RH BETWEEN FLANGE J1 & K								
219	69-52291-1		. SUPPORT ASSY, FIRE DETECTOR *[13]*[17]*[29]*[34]*[36]							A	1
219	69-52291-3		. SUPPORT ASSY, FIRE DETECTOR *[14]*[18]*[30]*[31]*[35]*[37]							B-FI-J	1
219	69-52291-5		. SUPPORT ASSY, FIRE DETECTOR *[32]							GH	1
220	BACS12CB3-7		. SCREW								4
221	NAS679A3W		. NUT								4
222	AN960-10L		. WASHER								4

- *[00] LIMITED USE ON 65-45799-1 AND 65-45798-1 BASIC INSTALLATIONS
- *[1] EACH ENGINE REQUIRES ONE 65-45799 PRATT AND WHITNEY BRACKET
INSTALLATION AND ONE 65-45798 BOEING BRACKET INSTALLATION.
- *[2] USED ONLY ON ENGINES WITH BOOM-TYPE VORTEX DISSIPATOR.
- *[3] NOT USED WHEN SB 73-1002 INCORPORATED
- *[4] DELETED.
- *[5] USED WITH 65-58135-17 TUBE ASSEMBLY AND 16234-1 HOSE ASSEMBLY (FIG. 1126).
- *[6] FOR INSTALLATION ON JT8D17 ENGINE ONLY.
- *[7] USED ON 69-53885-1.
- *[8] USED ON 65-45799-166.
- *[9] USED ON 65-45799-167.
- *[10] USED ON ENGINES WITH NOISE ATTENUATION TREATMENT.
- *[11] POST SB 80-1004 THRU 80-1008, 80-1011
- *[12] USED ON ENGINES WITHOUT NOISE ATTENUATION TREATMENT.
- *[13] PRE SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083,
71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1118, 71-1121, 71-1125,
71-1125R1, 71-1125R2, 71-1128, 71-1142, 71-1147, 71-1151, 71-1162, 71-1179, 71-1198
- *[14] POST SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083,
71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1118, 71-1121, 71-1125,
71-1125R1, 71-1125R2, 71-1128, 71-1142, 71-1147, 71-1151, 71-1162, 71-1179, 71-1198
- *[15] PRE SB 71-1102, 71-1111
- *[16] POST SB 71-1102, 71-1111
- *[17] PRE SB 71-1079, 71-1081, 71-1088, 71-1091, 71-1118
- *[18] POST SB 71-1058, 71-1079, 71-1081, 71-1088, 71-1091, 71-1118
- *[19] PRE SB 71-1058, 71-1114, 71-1079, 71-1081, 71-1088, 71-1091, 71-1118
- *[20] POST SB 71-1058, 71-1114, 71-1079, 71-1081, 71-1088, 71-1091, 71-1118
- *[21] PRE SB 71-1089
- *[22] POST SB 71-1089
- *[23] PRATT AND WHITNEY BRACKET SHOWN FOR REFERENCE. NOT PART OF 65-45798
OR 65-45799 INSTALLATIONS
- *[24] LIMITED USAGE ON 65-45799-147, -150, -170, -171
- *[25] PRE SB 30-1029
- *[26] POST SB 30-1029
- *[27] PRE SB 71-1094, -1156, -1085, -1162
- *[28] POST SB 71-1094, -1156, -1085, -1162
- *[29] PRE SB 71-1140
- *[30] POST SB 71-1140
- *[31] PRE SB 26-1025
- *[32] POST SB 26-1025
- *[33] POST SB 80-1045, 80-1045R1, 80-1050
- *[34] PRE SB 71-1182
- *[35] POST SB 71-1182
- *[36] PRE SB 71-1189, 71-1278
- *[37] POST SB 71-1189, 71-1278
- *[38] POST SB 80-1055
- *[39] PRE SB 71-1337
- *[40] POST SB 71-1337
- *[41] PRE SB 71-1340
- *[42] POST SB 71-1340
- *[43] PRE SB 71-1351
- *[44] POST SB 71-1351



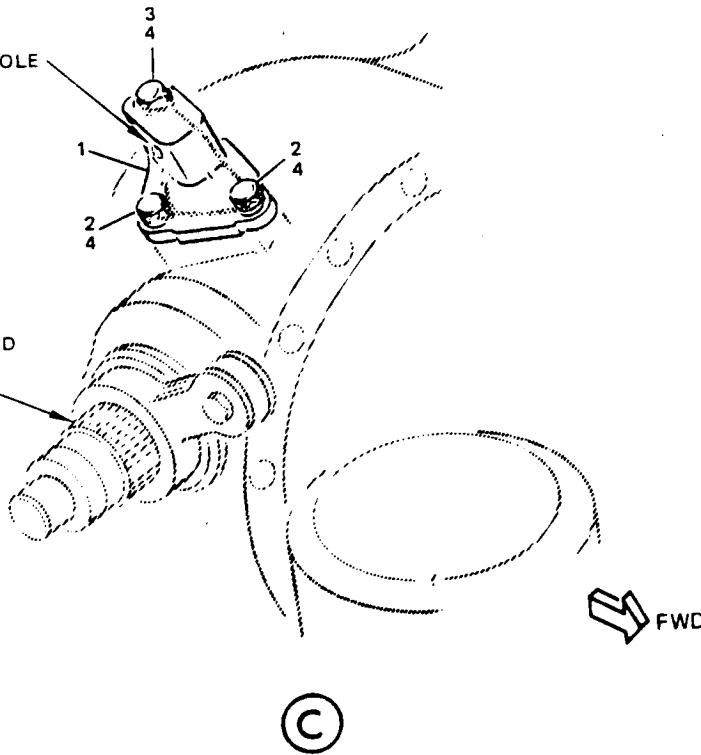
REF.
FIG.
536
548

Engine Control Linkage Installation
Figure 1110 (Sheet 1)



P & W POWER CONTROL
IDLE POSITION RIG PIN HOLE
(REF)

RIGHT HAND END
OF CONTROLS
CROSS SHAFT

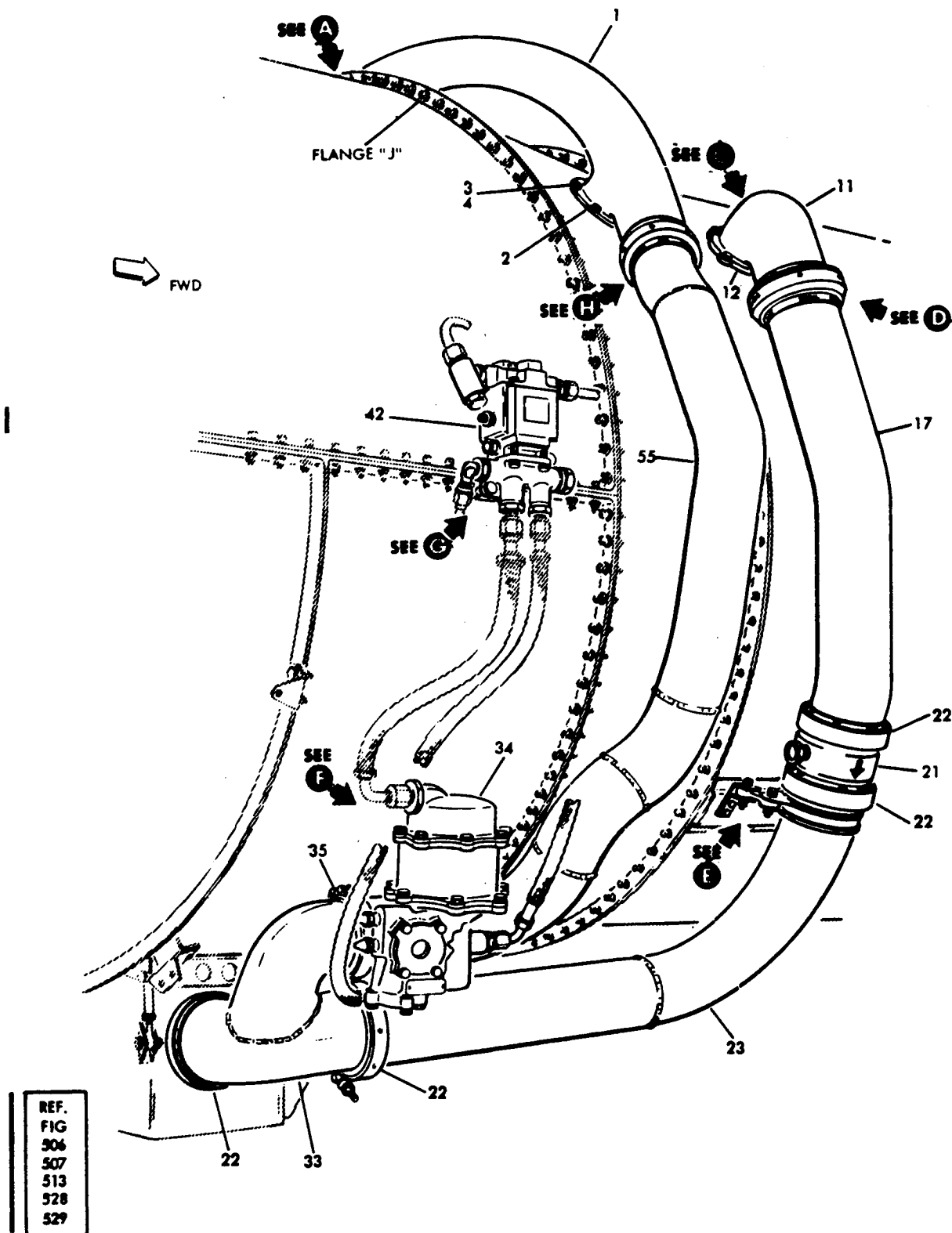


Engine Control Linkage Installation
Figure 1110 (Sheet 2)

FIG. & ITEM NO.	PART NO.	QEC SUB KIT NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	UNIT PER ASSY
				1	2	3	4	5	6	7		
1110-	65-51534-1			LINKAGE INSTL, ENGINE CONTROLS							A	RF
	65-51534-5			LINKAGE INSTL, ENGINE CONTROLS							B	RF
	65-51534-8			LINKAGE INSTL, ENGINE CONTROLS							C	RF
	65-51534-9			LINKAGE INSTL, ENGINE CONTROLS							D	RF
1	65-51517-3			. RIGGING PIN MOUNT								1
1	65-51517-1			. RIGGING PIN MOUNT (OPT)								1
2	NAS1304-4HW			. BOLT							AB	2
3	NAS1304-16HW			. BOLT							AB	1
4	AN960PD416L			. WASHER								3
5	65-50545-3			. CRANK, ENGINE START								1
5	65-50545-1			. CRANK, ENGINE START (OPT)							A	1
6	65-50546-1			. CRANK, ENGINE THRUST							AB	1
6A	65-51534-10			. CRANK ASSY (BONDED)							CD	1
6A	65-51534-11			. CRANK ASSY (BONDED) (OPT)							CD	1
6B	69-71542-1			. NUT (REPLS P&W NUT)							CD	1
7	BACB30NE4-11			. BOLT, (PREF) (USED ON 65-50546-1 WITH 69-66647-3 AND BACN10JC4)								1
7	BACB30NF4D11			. BOLT (OPT) (USED WITH BACN10JD104 AND MS24665-138)								1
7	NAS1104-11			. BOLT (OPT) (USED WITH BACN10JC4)								1
8	BACN10JC4			. NUT (USED WITH BACB30NE4-11 OR NAS1104-11)								1
8	BACN10JD104			. NUT (USED WITH BACB30NF4D11)								1
9	AN960C416			. WASHER								1
10	MS24665-138			. COTTER PIN (USED WITH BACB30NF4D11)								1
11	69-66647-3			. BOLT RETAINER (USED ON 65-50546-1)								1
12	BACB30NE5-11			. BOLT (PREF) (USED ON 65-50545-3 WITH 69-66647-8 AND BACN10JC5)								1
12	BACB30NF5D10			. BOLT (OPT) (USED WITH BACN10JD105 AND MS24665-138)								1
12	NAS1105-10			. BOLT (OPT) (USED WITH BACN10JC5)								1
13	BACN10JC5			. NUT (USED WITH BACB30NE5-11 OR NAS1105-10)								1
13	BACN10JD105			. NUT (USED WITH BACB30NF5D10)								1
14	AN960C516			. WASHER								1
15	MS24665-138			. COTTER PIN (USED WITH BACB30NF5D10)								1
16	69-66647-8			. BOLT RETAINER (USED ON 65-50545-1)								1
17	69-43481-2			. CLAMP HALF								2
18	BACN10JC4			. NUT								1
19	AN960C416			. WASHER								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1110-20	L6901		.	T	R	A	N	S	D	C	C	1
				(B	O	E	I	N	G	1	0-61988-1) *[00]
20	65-51534-12		.	T	R	A	N	S	D	C	C	1
21	NAS6604-16		.	B	O	L	T				C	1
22	AN960C416		.	W	A	S	H	E	R		C	3
23	BACN10JC4		.	N	U	T					C	2
24	BACB30NF4-13		.	B	O	L	T				C	1
25	BACB28AK04-030		.	B	U	S	H	I	N	G	C	2
26	69-71535-1		.	S	T	O	P				C	1

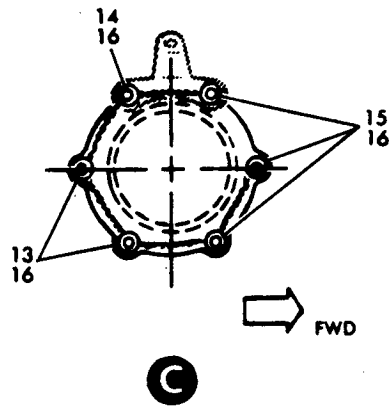
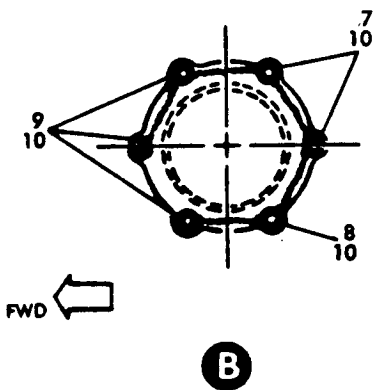
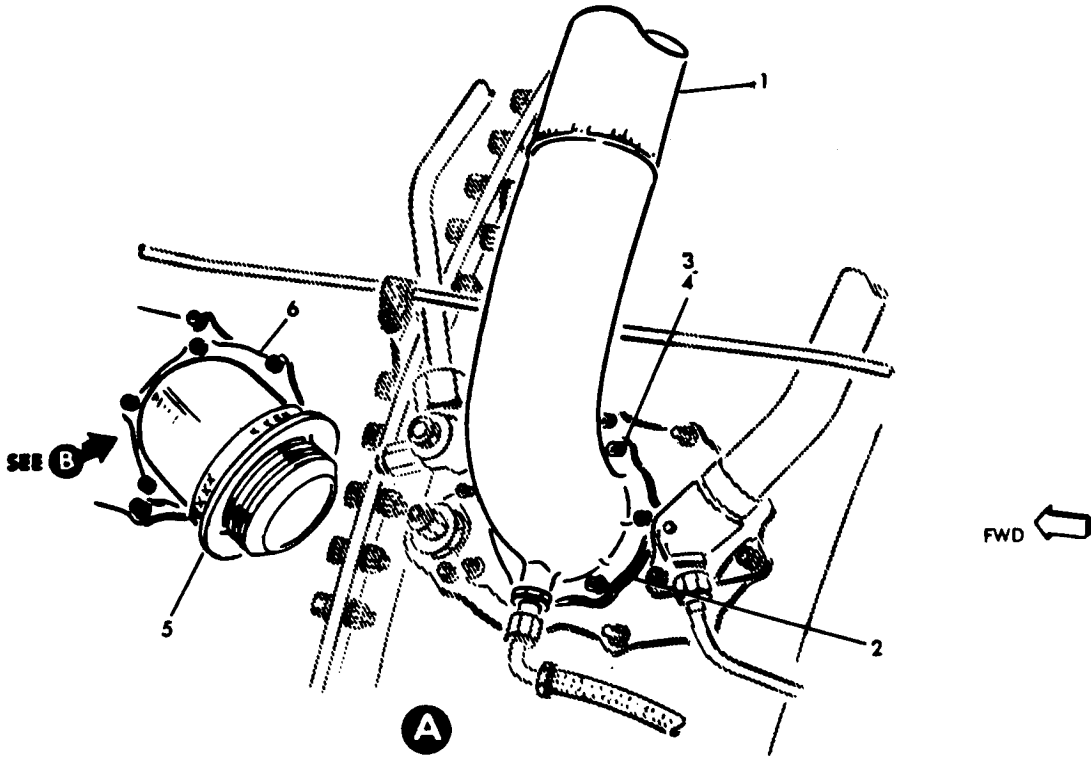
*[00] LIMITED USAGE



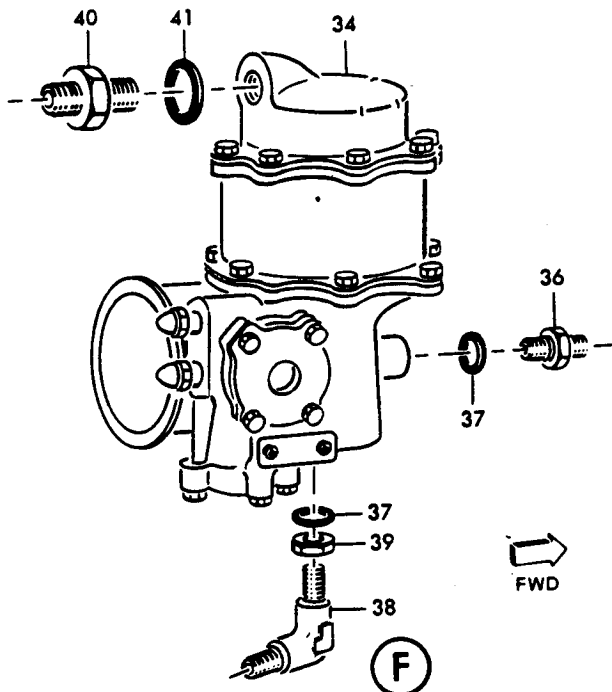
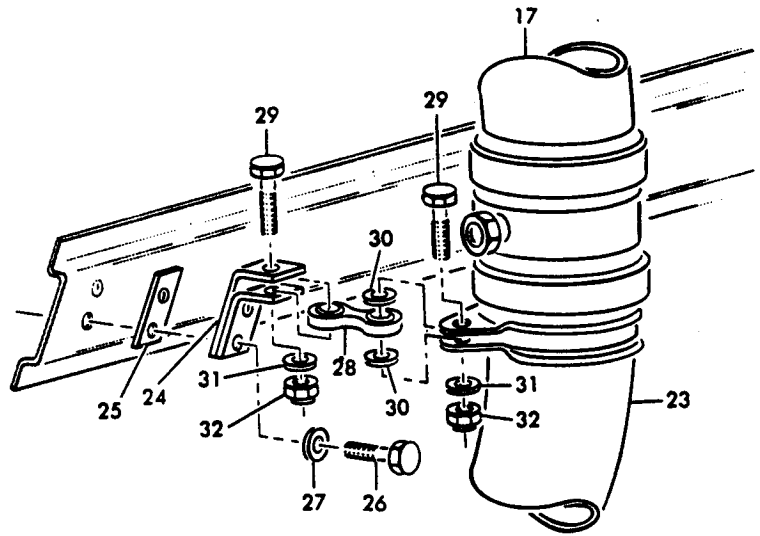
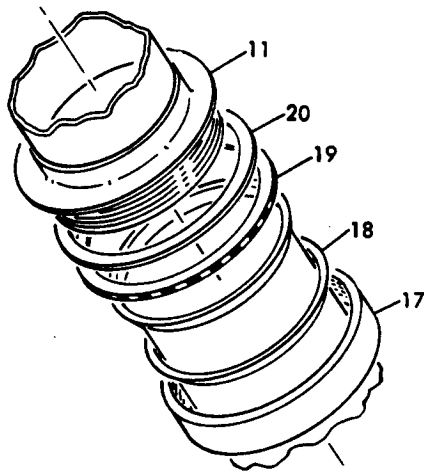
Duct Installation Air Conditioning Engines 1 & 2
Figure 1111 (Sheet 1)

May 15/68

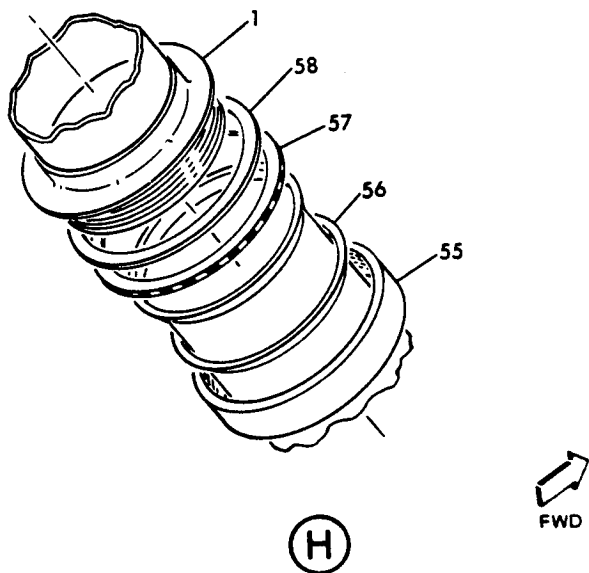
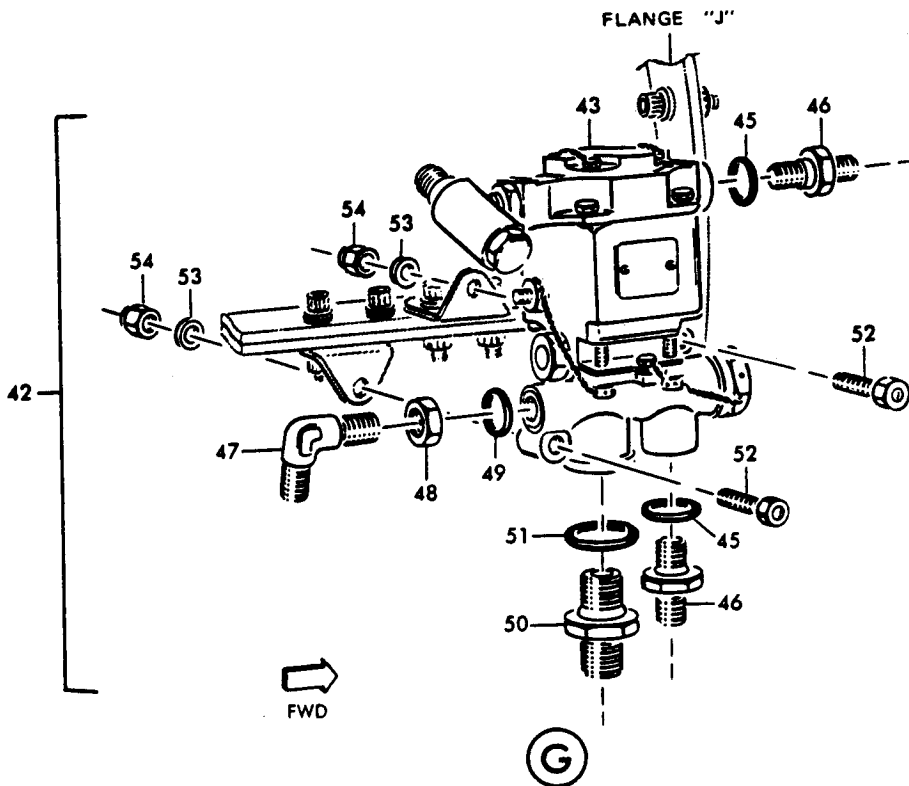
71-03-01
Page 1115



Duct Installation Air Conditioning Engines 1 & 2
Figure 1111 (Sheet 2)



Duct Installation Air Conditioning Engines 1 & 2
Figure 1111 (Sheet 3)



Duct Installation Air Conditioning Engines 1 & 2
 Figure 1111 (Sheet 4)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1111-	65-46851-1		DUCT INSTL, AIR CONDITIONING (ENG NO. 1, 2)							A	RF
	65-46851-2		DUCT INSTL, AIR CONDITIONING (ENG NO. 1, 2)							B	RF
	65-46851-3		DUCT INSTL, AIR CONDITIONING (ENG NO. 1, 2)							C	RF
	65-46851-4		DUCT INSTL, AIR CONDITIONING (ENG NO. 1, 2)							D	RF
	65-46851-5		DUCT INSTL, AIR CONDITIONING (ENG NO. 1, 2)							E	RF
1	65-22441-24		. MANIFOLD ASSY, HIGH PRESSURE (REPLS 65-22441-21 ON ENG RETROFITTED PER 65-75985)								1
1	65-22441-31		. MANIFOLD ASSY, HIGH PRESSURE *[4]*[7]							A	1
1	65-22441-28		. MANIFOLD ASSY, HIGH PRESSURE							BC	1
1	65-22441-34		. MANIFOLD ASSY, HIGH PRESSURE *[2]*[5]*[6]							D	1
1	65-22441-39		. MANIFOLD ASSY, HIGH PRESSURE*[3]							E	1
2	*[1]		. GASKET, V77445								2
3	BACB30EJ4H3		. BOLT								12
4	BACW10BP4CT		. WASHER								12
5	69-45175-1		. MANIFOLD ASSY, 8TH-STAGE LH								1
6	*[1]		. GASKET, V77445								1
7	BACB30EJ4H7		. BOLT								2
8	BACB30EJ4H3		. BOLT								1
9	BACB30EJ4H2		. BOLT								3
10	BACW10BP4CT		. WASHER								6
11	69-45174-1		. MANIFOLD ASSY, 8TH-STAGE RH								1
12	*[1]		. GASKET, V77445								1
13	BACB30EJ4H7		. BOLT								2
14	BACB30EJ4H3		. BOLT								1
15	BACB30EJ4H2		. BOLT								3
16	BACW10BP4CT		. WASHER								6
17	65-53865-1		. DUCT ASSY, 8TH-STAGE (OPT)								1
17	65-53865-11		. DUCT ASSY, 8TH-STAGE (OPT)								1
18	69-26229-2		. WEAR RING								1
18	69-64750-3		. WEAR RING (OPT)								1
19	69-56508-3		. SHIM, LAMINATED								1
20	69-56508-6		. SHIM, SOLID								1
21	123558-1-1		. VALVE, CHECK, V99193 (BOEING 10-60508-1,-3)								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1111-21	123558-1-3		.	VALVE, CHECK, V99193 (BOEING 10-605081-3)							1
21	123558-1-4		.	VALVE, CHECK, V99193 (BOEING 10-60508-3)							1
22	BAC10DU350Y		.	CLAMP (REPLS BACC10DU350C)							4
23	65-53865-802		.	DUCT ASSY							1
23	65-53865-803		.	DUCT ASSY (OPT)							1
24	69-43437-1		.	BRACKET ASSY							1
25	BACS40L12-28		.	SHIM, LAMINATED							1
26	NAS1104-6		.	BOLT							2
27	AN960C416L		.	WASHER							2
28	66-14598-1		.	LINK							1
29	NAS1104-8		.	BOLT							2
30	AN960C416L		.	WASHER							2
31	AN960C416		.	WASHER							2
32	BACN10JC4		.	NUT (REPLS NAS 679A4W)							2
33	65-53866-16		.	DUCT (PRFD)							1
33	65-53866-800		.	DUCT (OPT)							1
34	392716-1-1		.	VALVE, MODULATING AND SHUTOFF, V99193 (BOEING 10-60492-4)							1
35	BACC10DU300Y		.	CLAMP (REPLS BACC10DU300C)							2
36	MS21902-6		.	UNION							1
37	BACP11P-B6		.	GASKET							2
38	MS21908-6		.	ELBOW							1
39	AN924-6		.	NUT							1
40	MS21902-8		.	UNION							1
41	BACP11P-B8		.	GASKET							1
42	69-42802-1		.	REGULATOR INSTL, PRESSURE DIFFERENTIAL, ENG							1
43	392686-3-1		.	REGULATOR, PRESSURE DIFFERENTIAL, V99193							1
44	NOT USED										
45	BACP11PB6		.	GASKET							2
46	MS21902-6		.	UNION							2
47	MS21926-4		.	ELBOW							1
48	AN924-4		.	NUT							1
49	BACP11PB4		.	GASKET							1
50	MS21902-8		.	UNION							1
51	BACP11PB8		.	GASKET							1
52	NAS1104-15		.	BOLT							2
53	AN960-416L		.	WASHER							2

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1111-54	BACN10JC4										2
55	65-53864-800										1
56	69-26230-2										1
56	69-64750-2										1
57	69-56508-2										1
58	69-56508-5										1

*[1] REF PRATT AND WHITNEY PARTS CATALOG FOR PART NUMBER.

*[2] PRE SB 71-1102, 71-1111

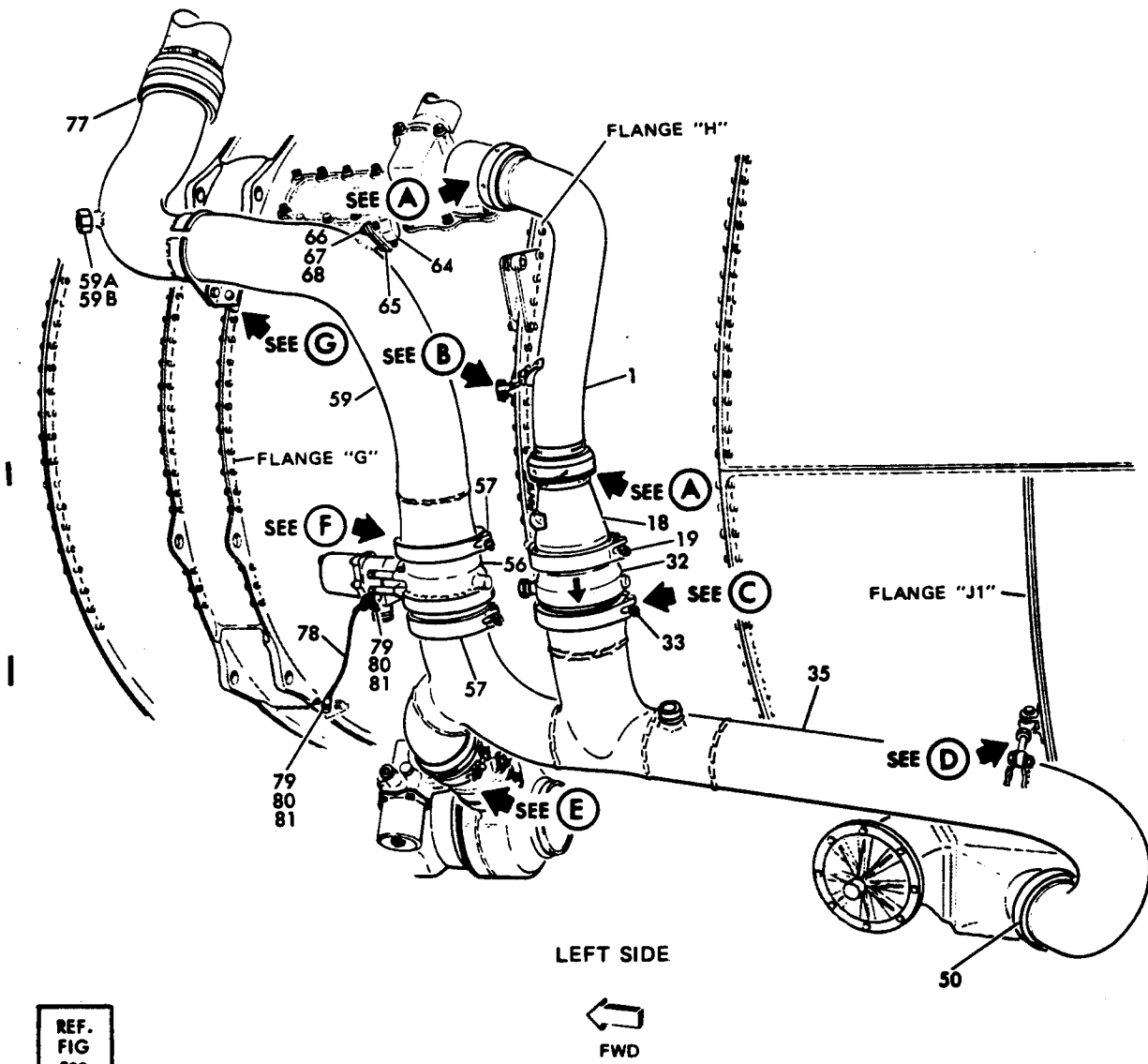
*[3] POST SB 71-1102, 71-1111

*[4] PRE SB 71-1055, 71-1058, 71-1086, 71-1135, 71-1080, 71-1088, 71-1081, 71-1138, 71-1118, 71-1154, 71-1137, 71-1087, 71-1162, 71-1182, 71-1192

*[5] POST SB 71-1055, 71-1058, 71-1086, 71-1135, 71-1080, 71-1088, 71-1081, 71-1138, 71-1118, 71-1154, 71-1137, 71-1087, 71-1162, 71-1182, 71-1192, 71-1258

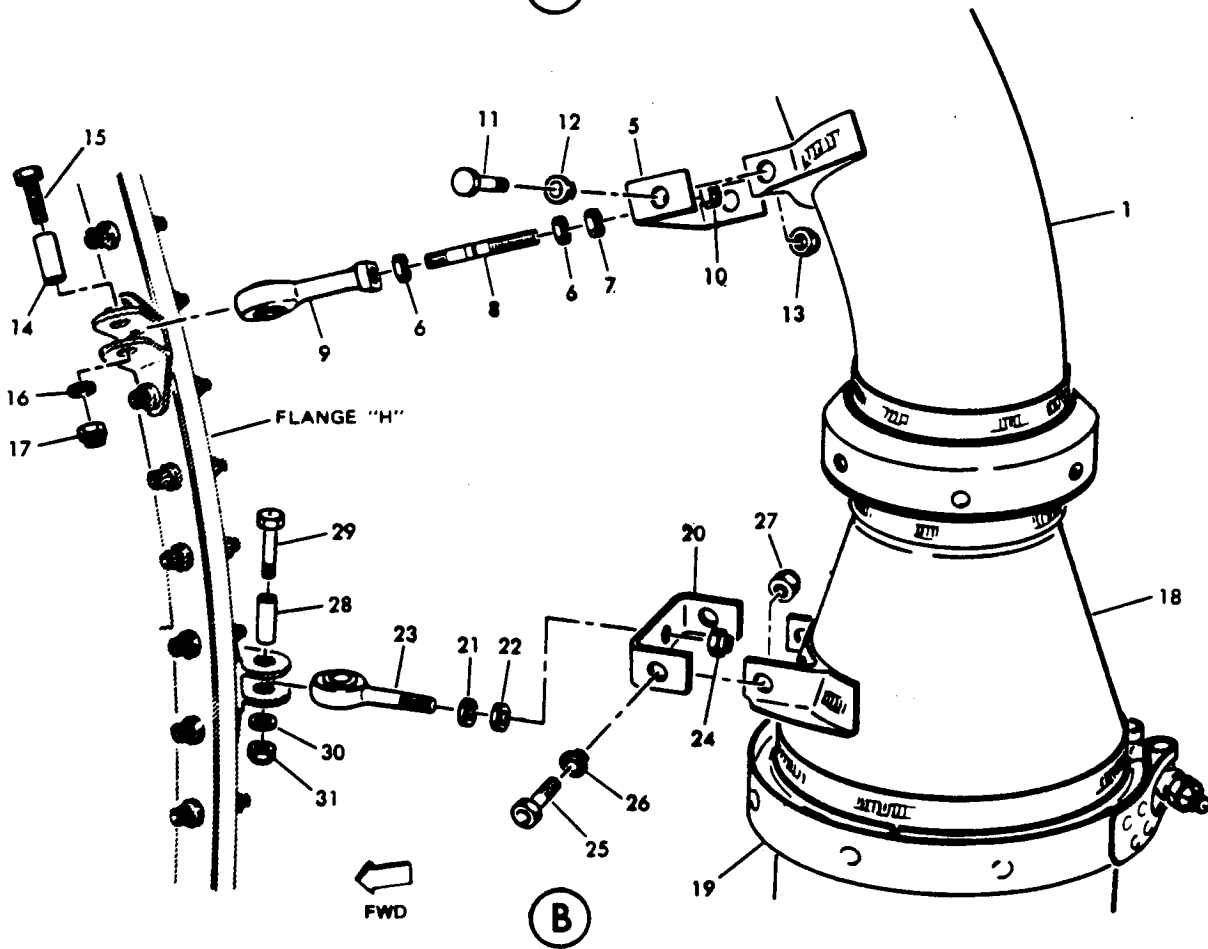
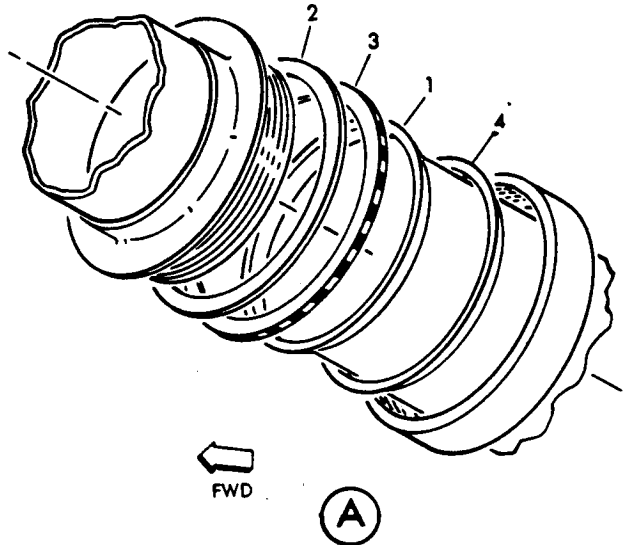
*[6] REPLACES 65-22441-21 PER MC 3510-193K

*[7] POST SB 71-1123, 71-1091, 71-1087

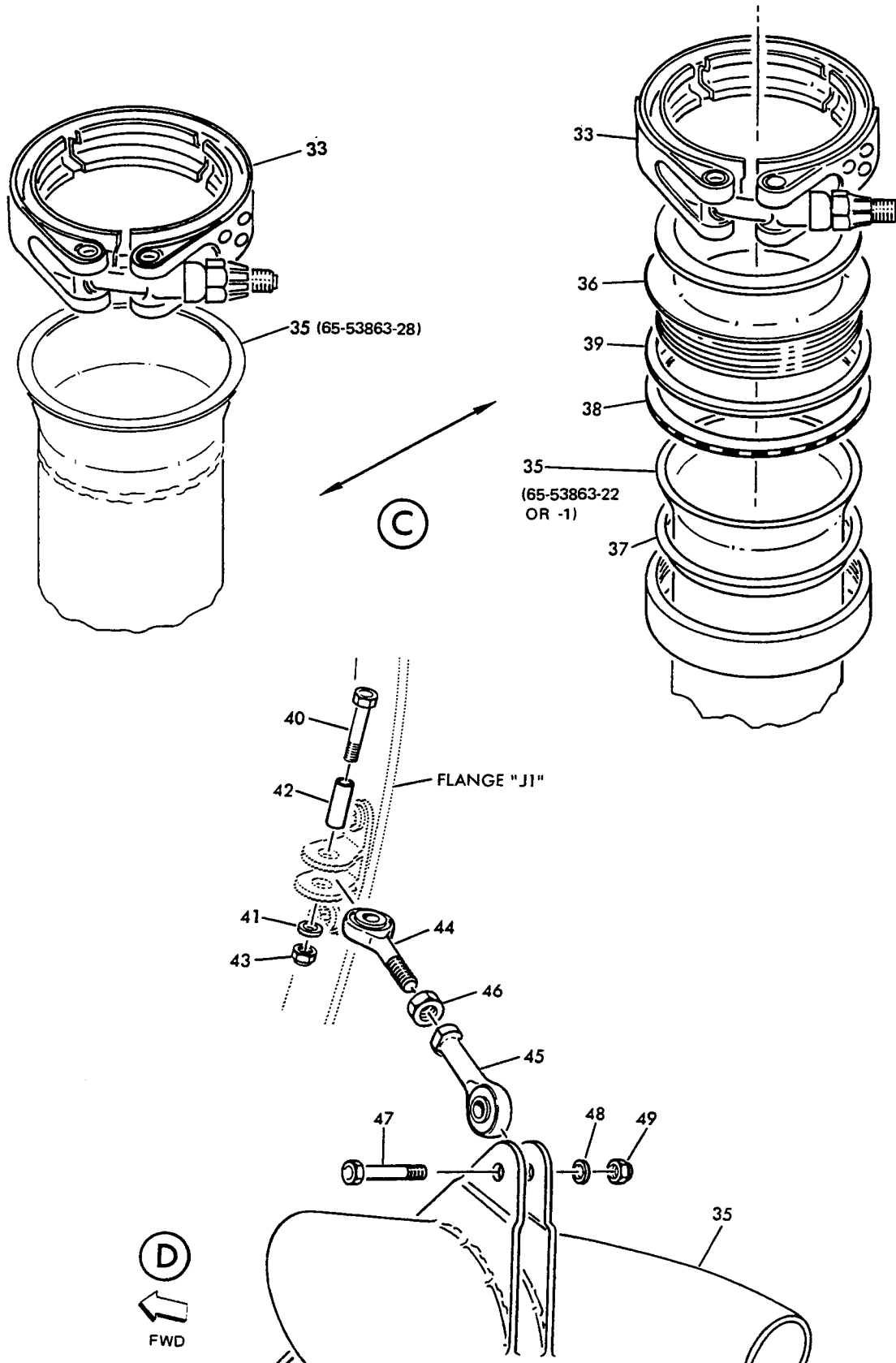


REF.
FIG
508
509
547

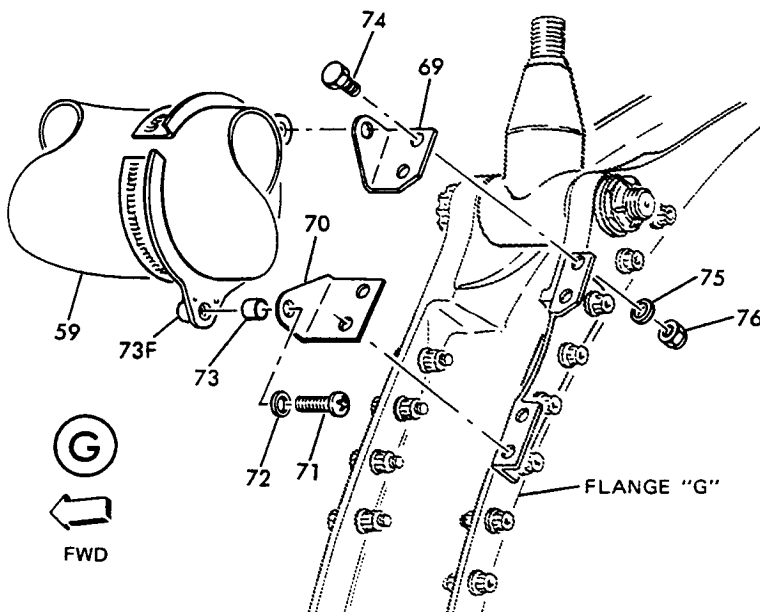
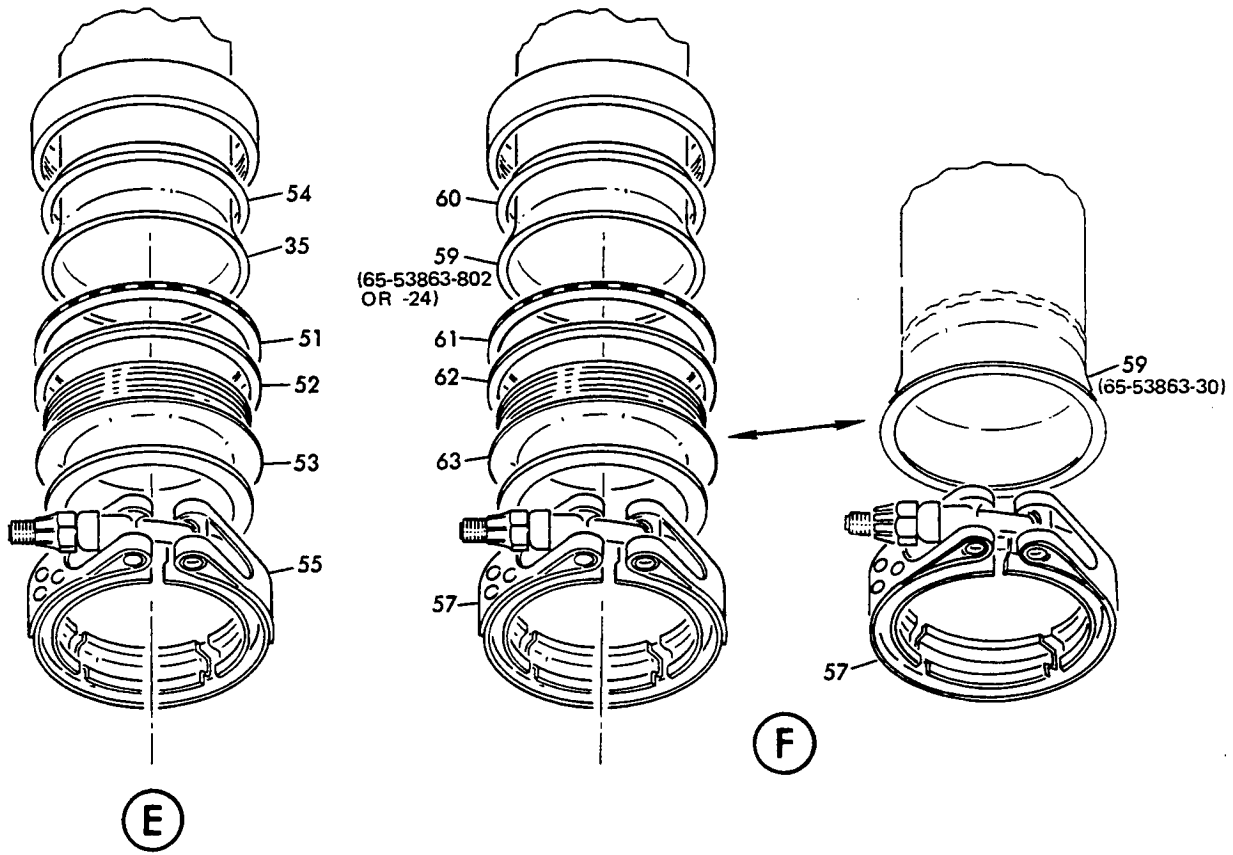
Cabin Air & Starter Ducts Installation
Figure 1112 (Sheet 1)



Cabin Air & Starter Ducts Installation
Figure 1112 (Sheet 2)



Cabin Air & Starter Ducts Installation
Figure 1112 (Sheet 3)



Cabin Air & Starter Ducts Installation
Figure 1112 (Sheet 4)

737

POWERPLANT BUILDUP



FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1112-	65-46852-1		DUCT INSTL - CABIN AIR AND STARTER							A	
	65-46852-2		DUCT INSTL - CABIN AIR AND STARTER							B	
1	69-44635-1		. DUCT ASSY								1
2	69-26231-2		. WEAR RING								2
2	69-64750-1		. WEAR RING (OPT)								2
3	69-56508-1		. SHIM, LAMINATED								2
4	69-56508-4		. SHIM, SOLID								2
5	69-44633-3		. BRACKET								1
6	AN316-4R		. NUT								2
7	AN315-4R		. NUT								1
8	NAS1454C4-0208		. ROD								1
9	BACB10J3		. ROD END								1
10	BACN10JC4C		. NUT (REPLS NAS679C4W)								2
11	NAS1103-3		. BOLT								2
12	BACB28X3C010		. BUSHING								2
13	BACN10JC3		. NUT (REPLS NAS679A3W)								2
14	NAS76-4-005P		. SPACER								1
15	NAS1104-11		. BOLT								1
16	AN960C416		. WASHER								1
17	BACN10JC4		. NUT (REPLS NAS679A4W)								1
18	69-44634-1		. DUCT ASSY (OPT)								1
18	69-44634-4		. DUCT ASSY (OPT)								1
19	BACC10DU350Y		. CLAMP (REPLS BAC10DU350C)								1
20	69-44633-3		. BRACKET								1
21	AN316-4R		. NUT								1
22	AN315-4R		. NUT								1
23	BACB10J12		. ROD END								1
24	BACN10JC4		. NUT (REPLS NAS679A4W)								1
25	NAS1103-3		. BOLT								2
26	BACB28X3C010		. BUSHING								2
27	BACN10JC3		. NUT (REPLS NAS679A3W)								2
28	NAS76A4-005P		. SPACER								1
29	NAS1104-11		. BOLT								1
30	AN960C416		. WASHER								1
31	BACN10JC4		. NUT (REPLS NAS679A4W)								1
32	123558-1-3		. VALVE, CHECK, V99193 (BOEING 10-60508-3)								1
32	123558-1-1		. VALVE, CHECK, V99193 (BOEING 10-60508-3)								1
33	BACC10DU350Y		. CLAMP (REPLS BACC10DU350C)								1

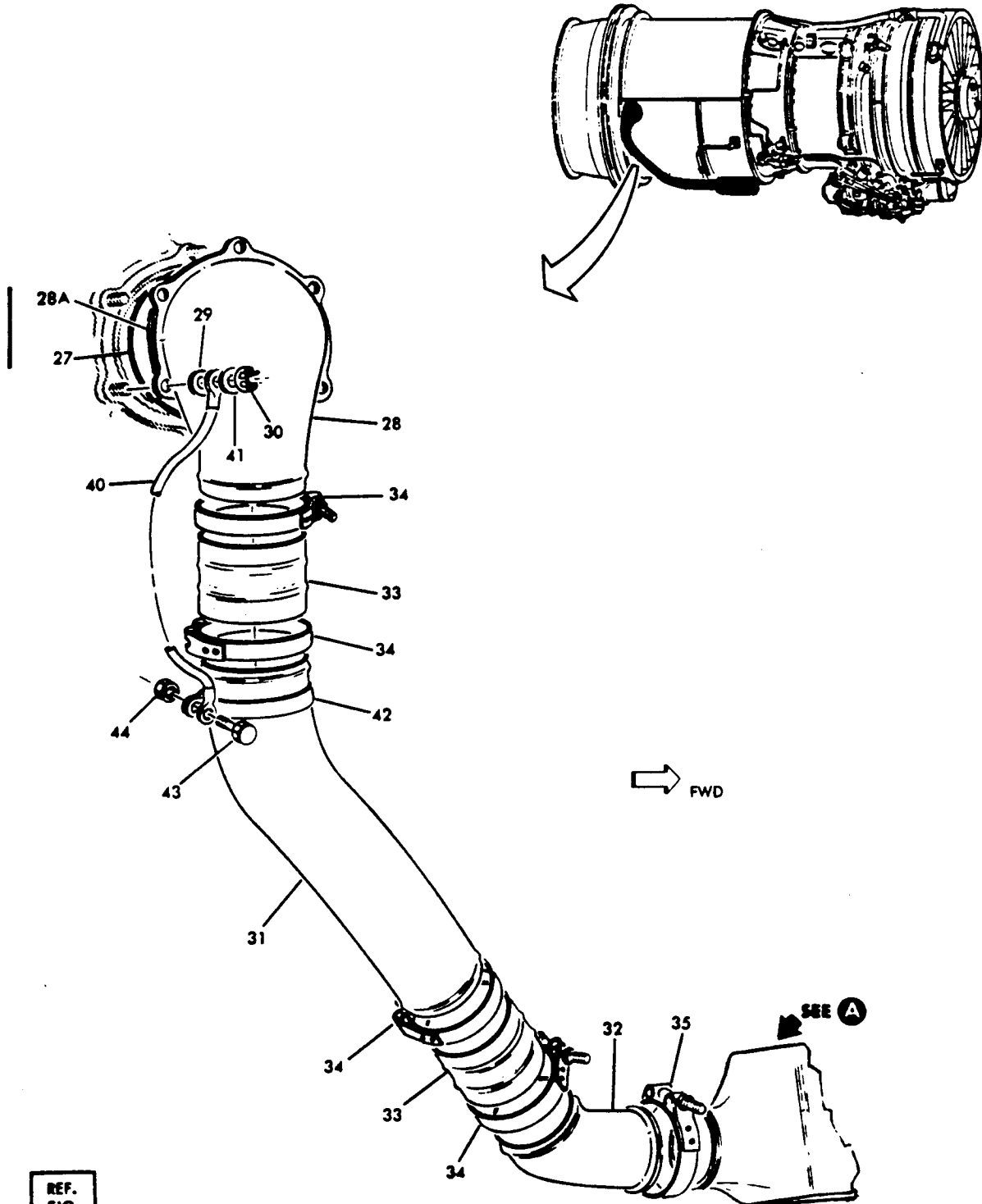
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1112-34	BACG10R381		DELETED								
35	65-53863-32		. DUCT ASSY (SB 36-1004)								1
35	65-53863-28		. DUCT ASSY (OPT)								1
35	65-53863-22		. DUCT ASSY (OPT)								1
35	65-53863-1		. DUCT ASSY (OPT)								1
36	69-17969-3		. RETAINER *[1]								1
37	69-26229-2		. WEAR RING *[1]								1
38	69-56508-3		. SHIM, LAMINATED *[1]								1
39	69-56508-6		. SHIM, SOLID *[1]								1
40	NAS1104-11		. BOLT								1
41	AN960C416		. WASHER								1
42	NAS76A4-005P		. SPACER								1
43	BACN10JC4		. NUT (REPLS NAS679A4W)								1
44	BACB10J12		. ROD END								1
45	BACB10A192M2		. ROD END								1
46	AN316-4R		. NUT								1
47	NAS1104-9		. BOLT								1
48	AN960-416		. WASHER								1
49	BACN10JC4		. NUT (REPLS NAS679A4W)								1
50	BACC10DU350Y		. CLAMP (REPLS BACC10DU350C)								1
51	69-56508-2		. SHIM, LAMINATED								1
52	69-56508-5		. SHIM, SOLID								1
53	69-20823-1		. RETAINER								1
54	69-26230-2		. WEAR RING, 3.0 IN. DIA								1
54	69-64750-2		. WEAR RING (OPT)								1
55	BACC10DP300A		. CLAMP								1
56	67-2905-001		. VALVE, SHUTOFF, V70138 (BOEING 10-60495-6)							A	1
56	2760000-101		. VALVE, SHUTOFF, V92003								1
57	BACC10DU350Y		. CLAMP (REPLS BACC10DU350C)								2
58	BACG10R381		DELETED								
59	65-53863-30		. DUCT ASSY								1
59	65-53863-802		. DUCT ASSY (OPT)								1
59	65-53863-24		. DUCT ASSY (OPT)								1
59A	BACP20AU8J		. PLUG (USED WITH 65-53863-24)								1
59B	AN901-8		. PACKING, O-RING (USED WITH 65-53863-24)								1
60	69-26229-2		. WEAR RING, 3.5 IN. DIA *[2]								1
60	69-64750-3		. WEAR RING (OPT)								1
61	69-56508-3		. SHIM, LAMINATED *[2] (REPLS 69-17963-6)								1
62	69-56508-6		. SHIM, SOLID *[2] (REPLS 69-17963-9)								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY																		
			1	2	3	4	5	6	7																				
1112-63	69-17969-3		.	R	E	T	A	I	N	E	R	*	[2	1														
64	129240-1-1		.	V	A	L	V	E	,	T	H	E	R	M	O	S	T	A	T	V	9	9	1	3					
65	3163408-1		.	G	A	S	K	E	T	,	V	9	9	1	9	3	1	1											
65	856411-1		.	G	A	S	K	E	T	,	V	9	9	1	9	1	9	1	(O	P	T)						
66	NAS1103-8		.	B	O	L	T																	2					
67	AN960-10L		.	W	A	S	H	E	R																2				
68	BACN10JC3		.	N	U	T		(R	E	P	L	S	N	A	S	6	7	9	A	3	W)			2			
69	69-43424-6		.	C	L	I	P		(O	P	T)													1			
69	69-43424-4		.	C	L	I	P		(O	P	T)	(S	B	7	1	-	1	0	1	1)			1		
69	69-43424-8		.	C	L	I	P		(O	P	T)	(P	R	E	S	B	3	6	-	1	0	1	3)	1	
69	69-43424-10		.	C	L	I	P		(P	O	S	T	S	B	3	6	-	1	0	1	3)			1			
70	69-43424-5		.	C	L	I	P		(O	P	T)														1		
70	69-43424-3		.	C	L	I	P		(O	P	T)	(S	B	7	1	-	1	0	1	1)			1		
70	69-43424-7		.	C	L	I	P		(O	P	T)	(P	R	E	S	B	3	6	-	1	0	1	3)	1	
70	69-43424-9		.	C	L	I	P		(P	O	S	T	S	B	3	6	-	1	0	1	3)			1			
71	NAS603-2		.	S	C	R	E	W		(P	R	E	S	B	3	6	-	1	0	1	3)			2			
71	NAS6603-9		.	S	C	R	E	W		(P	O	S	T	S	B	3	6	-	1	0	1	3)			2		
72	AN960C10		.	W	A	S	H	E	R		(P	R	E	S	B	3	6	-	1	0	1	3)			2		
72	AN960C10L		.	W	A	S	H	E	R		(P	O	S	T	S	B	3	6	-	1	0	1	3)			2	
73	NAS43HT3-8		.	S	P	A	C	E	R		(P	R	E	S	B	3	6	-	1	0	1	3)			2		
73	NAS43HT3-28		.	S	P	A	C	E	R		(P	O	S	T	S	B	3	6	-	1	0	1	3)			2	
73F	BACN10LHC38K		.	N	U	T	P	L	A	T	E															2			
73G	MS20427M3		.	R	I	V	E	T																		4			
74	NAS1103-1		.	B	O	L	T		(P	R	E	S	B	3	6	-	1	0	1	3)					4		
74	BACB30LM3U2		.	B	O	L	T		(P	O	S	T	S	B	3	6	-	1	0	1	3)				4		
75	AN960-10L		.	W	A	S	H	E	R		(P	R	E	S	B	3	6	-	1	0	1	3)			4		
75	AN960C10L		.	W	A	S	H	E	R		(P	O	S	T	S	B	3	6	-	1	0	1	3)			4	
76	BACN10JC3		.	N	U	T		(R	E	P	L	S	N	A	S	6	7	9	A	3	W)			4			
76	BACN10YR3C		.	N	U	T		(P	O	S	T	S	B	3	6	-	1	0	1	3)				4			
77	BACC10DU350Y		.	C	L	A	M	P		(R	E	P	L	S	B	A	C	C	1	0	D	U	3	5	0	C)	1
78	MS25083-2BB10		.	J	U	M	P	E	R	A	S	S														1			
79	BACS12CB3-8		.	S	C	R	E	W																		2			
80	AN960-10		.	W	A	S	H	E	R																		2		
81	BACN10JC3		.	N	U	T		(R	E	P	L	S	N	A	S	6	7	9	A	3)					1		

*[1] USED WITH 65-63863-1, -22

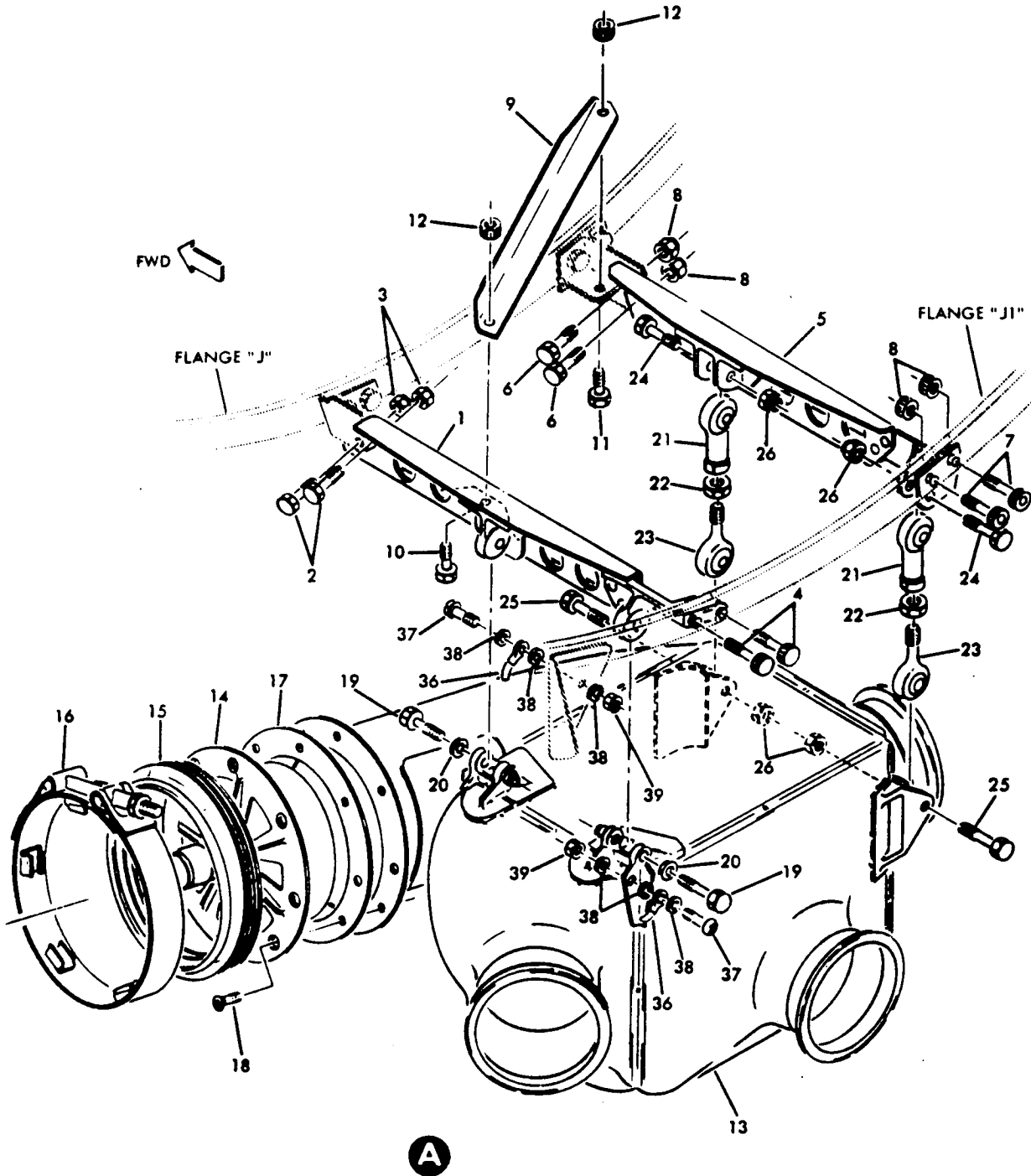
*[2] USED WITH 65-53863-24, -802

*[3] USED ON BRACKET ASSY 65-43424-1 WHICH IS WELDED TO DUCT ASSY 65-53863-30



REF.
FIG
510
527

Engine Air Conditioning System Pre-cooler Installation
Figure 1113 (Sheet 1)



Engine Air Conditioning System Precooler Installation
Figure 1113 (Sheet 2)

Jun 10/72

737

POWERPLANT BUILDUP



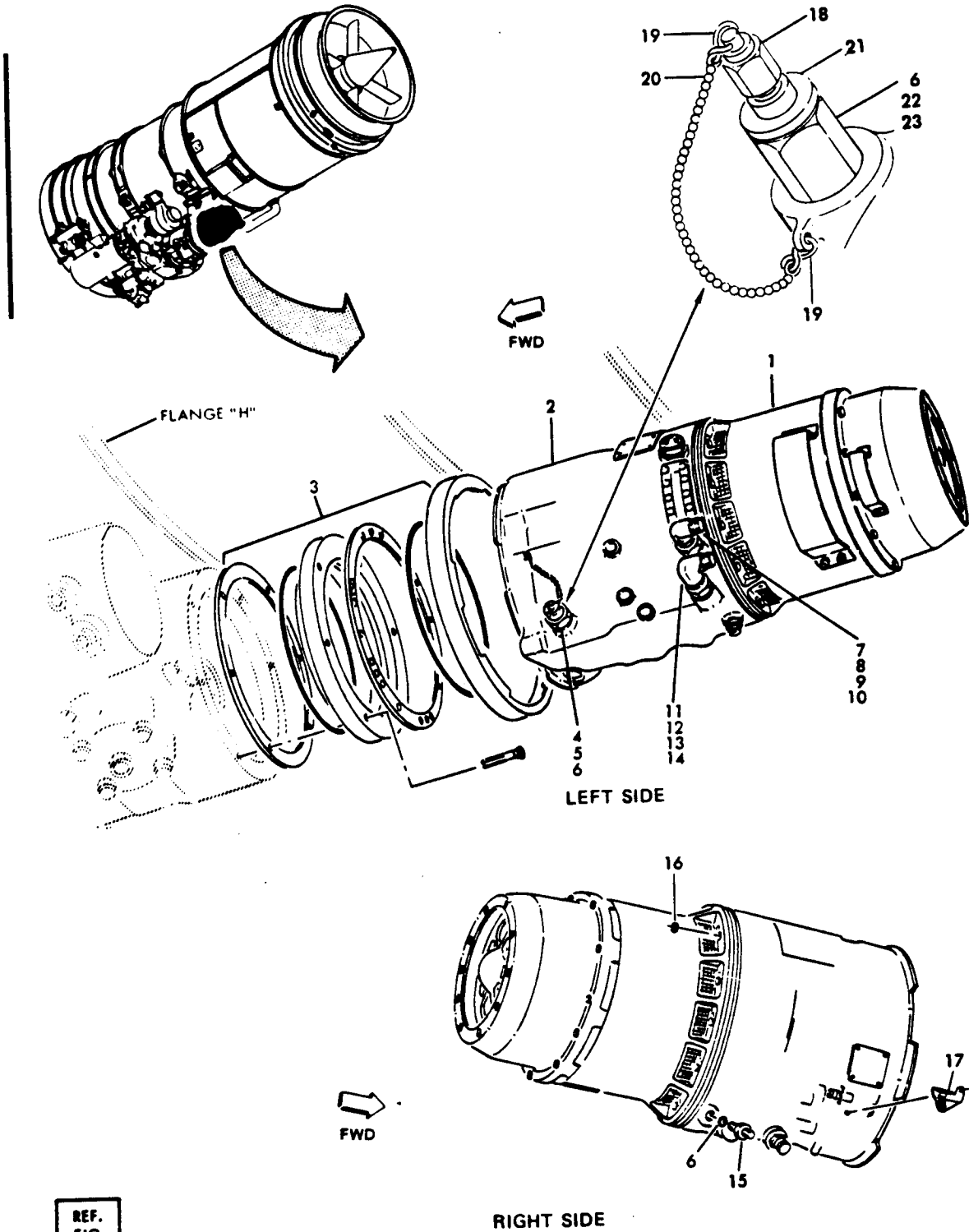
OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1113-	65-46853-1		PRECOOLER INSTL, ENGINE AIR CONDITIONING SYSTEM							A	RF
	65-46853-2		PRECOOLER INSTL, ENGINE AIR CONDITIONING SYSTEM							B	RF
1	69-42570-1		. SUPPORT ASSY, PRECOOLER *[4]								1
1	69-42570-2		. SUPPORT ASSY, PRECOOLER *[4]								1
2	NAS1304-2		. BOLT								2
3	BACN10BL4L		. NUT								2
4	BACB30CW4H2		. BOLT								2
5	69-42571-1		. SUPPORT ASSY, PRECOOLER *[4]								1
5	69-42571-2		. SUPPORT ASSY, PRECOOLER *[4]								1
6	NAS1304-3		. BOLT								2
7	BACB30CW4-4		. BOLT								2
8	BACN10BL4L		. NUT								4
9	69-42569-3		. ANGLE, PRECOOLER SUPPORT								1
10	NAS1303-1		. BOLT								1
11	NAS1303-2		. BOLT								1
12	NAS679C3W		. NUT								2
13	182400-1-1		. PRECOOLER, V70210								1
14	392664-1-1		. REGULATOR ASSY, V99193								1
15	D58364		. SEAL ASSY, V16519								1
15	69-46654-1		. SEAL ASSY (OPT)								1
16	69-46653-1		. LOCK RING ASSY								1
17	69-42533-1		. GASKET								1
18	NAS517-2-1		. SCREW								8
19	69-42825-1		. BOLT								2
20	MS20002C4		. WASHER								2
21	BACB10A180D		. BEARING								2
22	AN316C5R		. NUT								2
23	BACB10A187		. BEARING								2
24	NAS1104-11		. BOLT								2
25	NAS1304-9		. BOLT								2
26	NAS679C4W		. NUT								4
27	377129		. SEAL, V77445								1
27	MS9386-161		. SEAL (OPT)								1
28	65-50889-1		. ADAPTER								1
28A	69-62274-1		. SPACER (USED ON 65-46853-2) *[2]							B	1
28A	69-62274-1		. SPACER *[3]							A	1
29	AN960PD416		. WASHER								6

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1113-											
30	BACN10BL4L		.							6	
31	69-62261-1		.						B	1	
31	65-53867-4		.						A	1	
31	65-53867-1		.						A	1	
31	69-62261-1		.						A	1	
32	65-53867-2		.							1	
33	BACD40D14-12-36		.							2	
33	BACD40D14-12-23		.							2	
34	BACC10BN362LR		.							4	
35	BACC10DU350AB		.							1	
35	BACC10DU350Y		.							1	
35	BACC10DU350C		.							1	
36	MS25083-2BB4		.							1	
37	NAS623-3-3		.							2	
38	AN960-10L		.							6	
39	NAS679A3W		.							2	
40	MS25083-2BC8		.							1	
41	AN960PD416		.							1	
42	AN735D56		.							1	
43	BACS12CB3-6		.							1	
44	NAS679A3W		.							1	

- *[1] PRE SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1081, 71-1083, 71-1088, 71-1091, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151, 71-1162, 71-1179, 71-1182, 71-1198, 71-1278
- *[2] POST SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1081, 71-1083, 71-1088, 71-1091, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1142, 71-1147
- *[3] POST SB'S 71-1032, 71-1055, 71-1079, 71-1140, 71-1151, 71-1162, 71-1179, 71-1182, 71-1198, 71-1278
- *[4] LIMITED USAGE

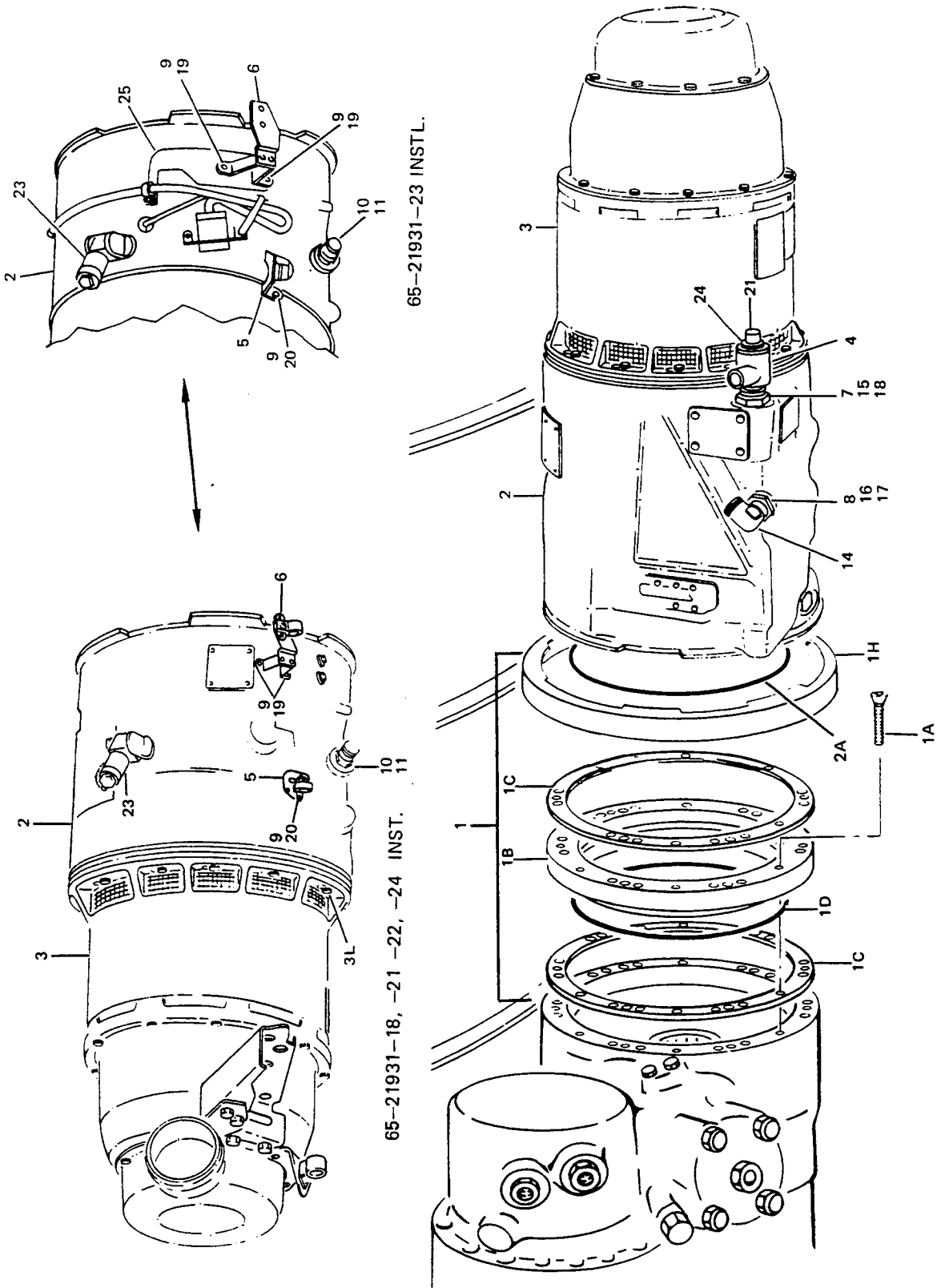


REF.
 FIG
 501
 522

AC Generator and Constant Speed Drive Installation
 Figure 1114

Aug 15/69

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1114-	65-50571-1		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL							A	RF
	65-50571-2		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL							B	RF
	65-50571-4		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL							C	RF
	65-50571-6		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL							D	RF
	65-50571-7		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL							E	RF
1	976J498-2		. AC GENERATOR, 40 KVA V83843 (BOEING 10-61224-12)(PREF)							ABCE	1
1	976J598-1		. AC GENERATOR, 40 KVA, V83843 (BOEING 10-61224-10)(OPT)							ABCE	1
1	976J498-1		. AC GENERATOR, 40 KVA, V83843 (BOEING 10-61224-2)(OPT)							ABCE	1
1	976J498-1		. AC GENERATOR, 40 KVA, V83843 (BOEING 10-61224-2)							D	1
2	699647		. CONSTANT SPEED DRIVE ASSY, V99167 (BOEING 10-61224-2)								1
3	689460A		. QAD ADAPTER ASSY, V99167 (BOEING 10-60295-15)								1
4	OMP2506-2		. COUPLING HALF, V86020							ACDE	1
5	OMP2506-4		. DUST COVER, V86020							ACDE	1
6	BACP11P112		. O-RING								2
7	MS21908-10		. ELBOW								1
8	AN6289-10		. NUT, JAM								1
9	MS28777-10		. RING, BACKUP								1
10	BACP11PB10		. O-RING								1
11	MS21908-8		. ELBOW								1
12	AN6289-8		. NUT, JAM								1
13	MS28777-8		. RING, BACKUP								1
14	BACP11PB8		. O-RING								1
15	MS21902-6		. UNION								1
16	BACN10GW6		. NUT (PREF)								12
16	BACN10BL6L		. NUT (OPT)								12
16	624AG5HD		. NUT (OPT)(V29372)								12
17	69-52217-1		. BRACKET ASSY								1
18	AN929A3		. CAP ASSY, PRESSURE SEAL							B	1
19	NAS1090-1		. HOOK							B	2
20	NAS1201CA8A		. CHAIN, BEAD							B	1
21	1611-0716		. SEAL WASHER, V99167							B	1
22	694282		. CHECK VALVE ASSY, V99167							B	1
23	699771		. ADAPTER ASSY, V99167							B	1

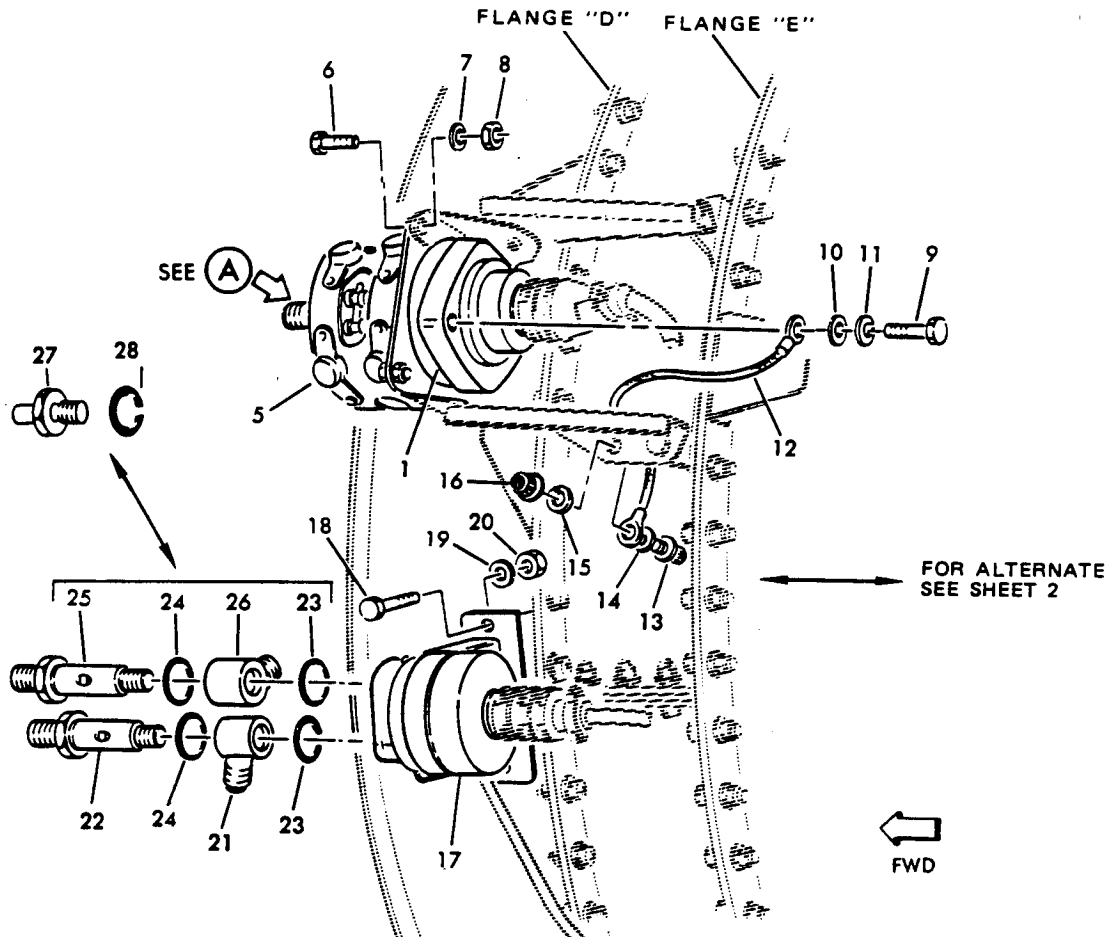
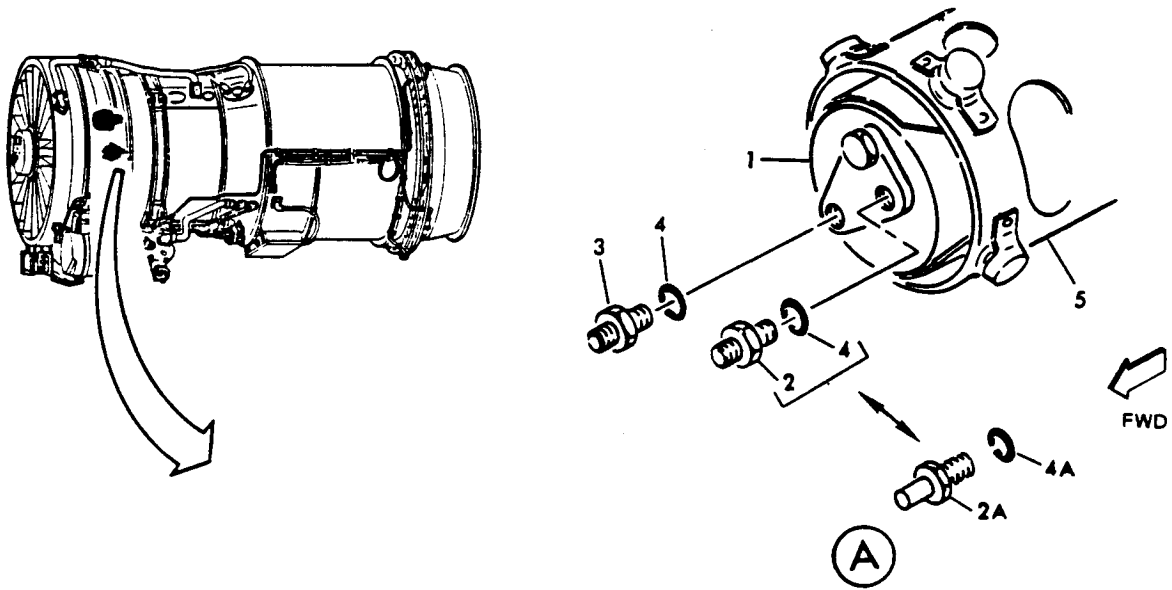


AC Generator and Constant Speed Drive Installation
Figure 1114A

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1114A-											
	65-21931-18		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL						A	RF	
	65-21931-21		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL						B	RF	
	65-21931-22		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL						C	RF	
	65-21931-23		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL						D	RF	
	65-21931-24		AC GENERATOR AND CONSTANT SPEED DRIVE INSTL						E	RF	
1	689460A		. QAD ADAPTER ASSY, V99167 (BOEING 10-60295-15)							1	
1A	0642C624-20		. . SCREW, FLAT SOCKET HD CAP, V99167							12	
1B	696692		. . PLATE, ADAPTER, V99167							1	
1C	693206		. . GASKET, V99167							2	
1D	M83248-1-176		. . PACKING, V99167							1	
-1E	693608		. . RING ASSY, V99167							1	
-1F	1802-24-2		. . . PIN, V99167							1	
-1F	MS16562-198		. . . PIN (OPT), V99167							1	
-1G	686751		. . . NUT, V99167							1	
1H	693601		. . . RING, V99167							1	
2	700842		DELETED							1	
2	731753		.CONSTANT SPEED DRIVE ASSY, V99167 (BOEING 10-61066-10)						ABCE	1	
2	735511A		.CONSTANT SPEED DRIVE ASSY, V99167 (BOEING 10-61066-11)						D	1	
2A	M83248/1-270		. . O-RING, V99167 *[1]							1	
3	976J498-2		.AC GENERATOR, V83843 (BOEING 10-61224-12)						A-D	1	
3	976J498-1		. AC GENERATOR, V83843 (BOEING 10-61224-2) (OPT)							1	
3	976J449-1		. AC GENERATOR, V83843 (BOEING 10-61224-2) (OPT)							1	
3	976J598-1		. AC GENERATOR, V83843 (BOEING 10-61224-10) (OPT)						A-D	1	
-3A	908C828-1		. . COVER, TERMINAL BOARD, V83843							1	
-3B	P28B9560-3		. . SCREW							2	
-3C	AN960-8		. . WASHER							2	
-3D	918B105-2		. . NUT, V83843							4	
-3E	MS35338-43		. . WASHER							4	
-3F	AN960-10L		. . WASHER							4	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1114A-											
-3G	908D049-1		.	.	COVER, TERNAL, V83843						1
-3H	918B105-4		.	.	NUT, V83843						6
-3I	MS35338-46		.	.	WASHER						6
-3J	AN960-616		.	.	WASHER						6
-3K	9448791-1		.	.	WASHER						6
3L	BACN10BL6L		.		NUT						12
4	69-71573-1		.		BULKHEAD FITTING						1
5	69-71727-1		.		BRACKET ASSY				ACE		1
5	69-71727-9		.		BRACKET ASSY				B		1
5	69-71727-10		.		BRACKET ASSY				D		1
6	69-71727-2		.		BRACKET ASSY						1
7	AN6289-10		.		NUT-JAM						1
8	AN6289-8		.		NUT-JAM						1
9	AN960C10L		.		WASHER						4
10	BACP11PB6		.		O-RING						1
11	BACU24K6		.		UNION						1
12	MS20995NC32				DELETED						
13	MS21902-6		.		UNION						1
14	MS21908-8		.		ELBOW						1
15	MS28773-10		.		WASHER						1
16	MS28773-8		.		WASHER						1
17	M83248-2-908		.		O-RING						1
18	M83248-2-910		.		O-RING						1
19	NAS1352C3-10		.		SCREW, CAP						2
20	NAS1352C3-8		.		SCREW, CAP						2
21	55584		.		THERMAL SWITCH, V99167						1
22	730019				DELETED						
23	706183		.		CAP, VENTED, V99167						1
24	M83248-2-908		.		O-RING						1
25	69-74656-1		.		PLATE, COVER				D		1

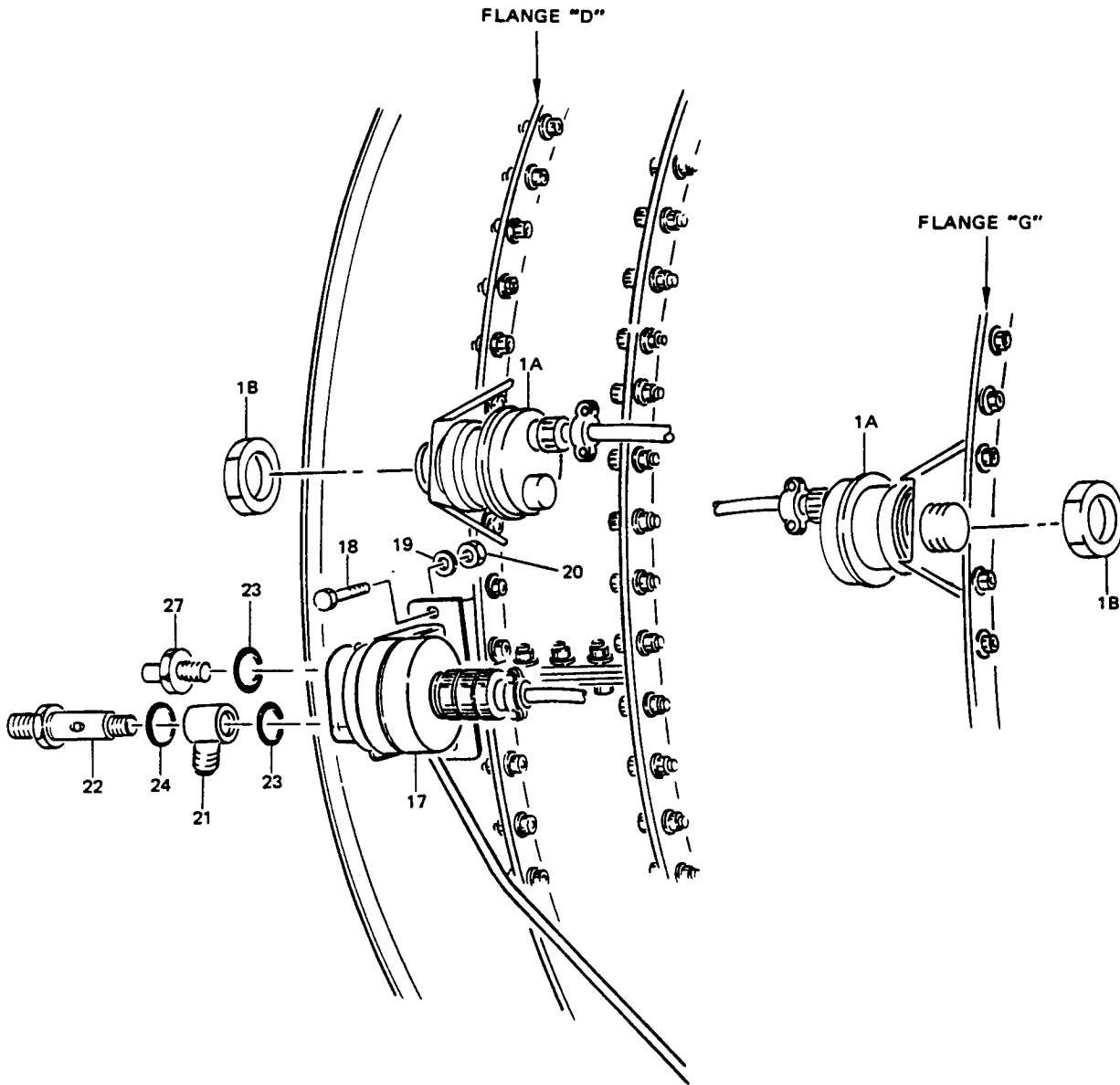
*[1] ITEM SUPPLIED WITH CONSTANT SPEED DRIVE ASSY



REF.
FIG.
505
543

65-50882-1, -2 INSTALLATION

Engine Oil Pressure and Low Pressure Warning
Transmitter and Switch Installation
Figure 1115 (Sheet 1)



65-50882-3 INSTALLATION

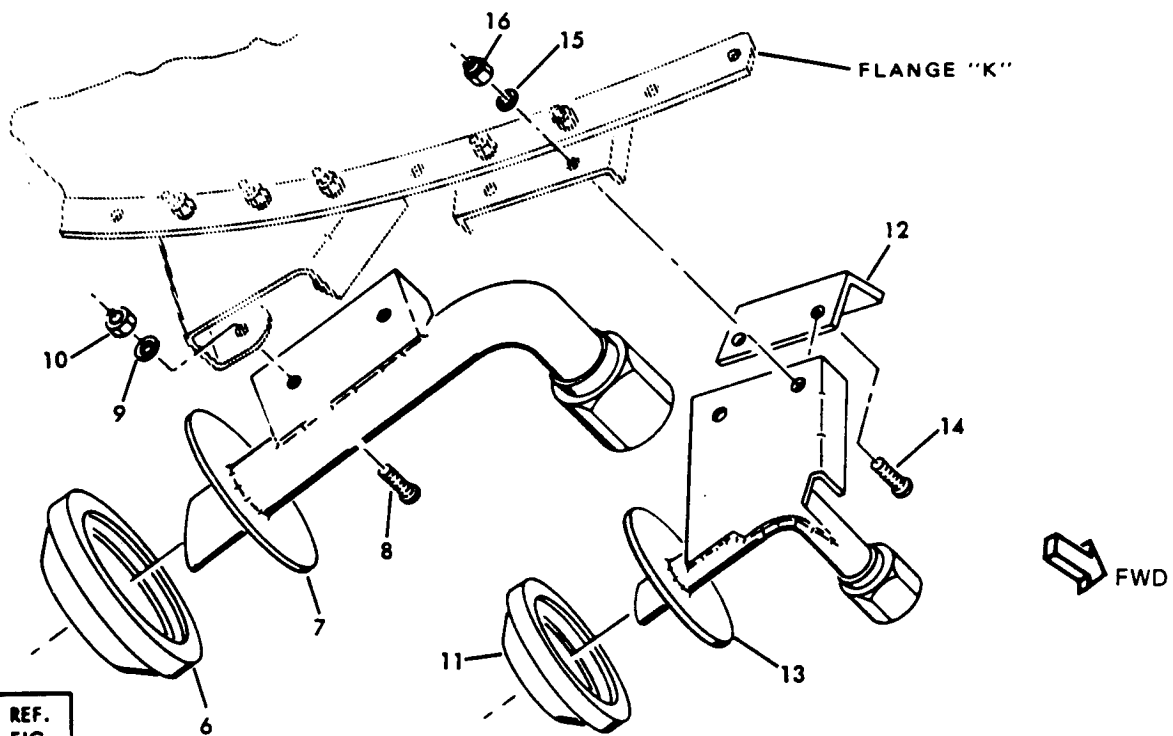
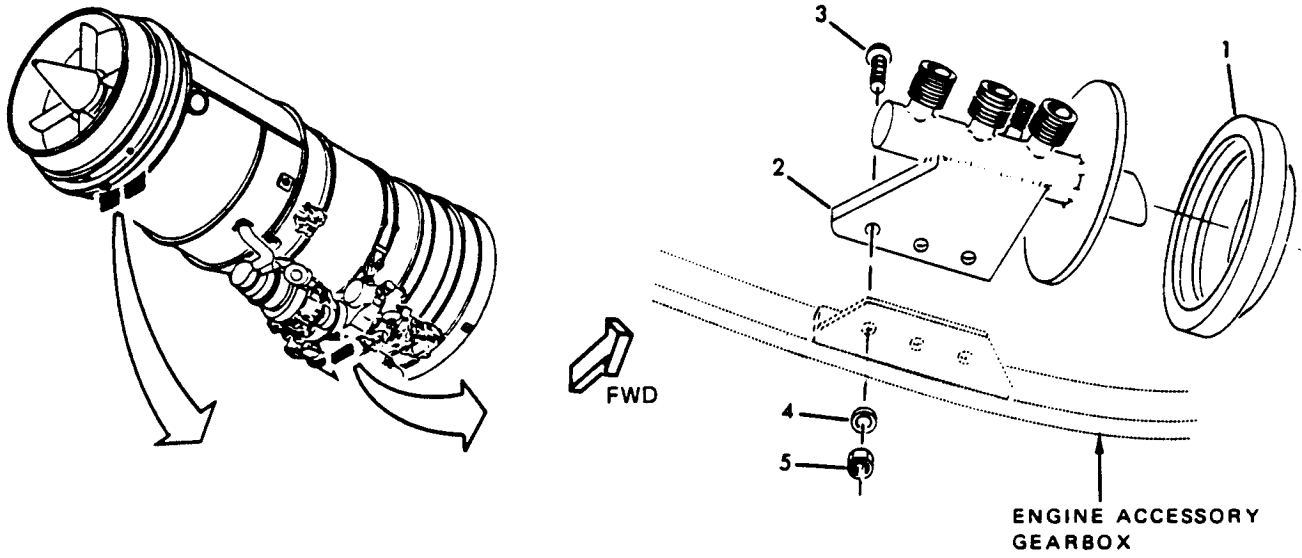
Engine Oil Pressure and Low Pressure Warning
Transmitter and Switch Installation
Figure 1115 (Sheet 2)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1115-	65-50882-1		TRANSMITTER AND SWITCH INSTL, ENG OIL PRESSURE AND LOW PRESSURE WARNING							A	RF
	65-50882-2		TRANSMITTER AND SWITCH INSTL, ENG OIL PRESSURE AND LOW PRESSURE WARNING							B	RF
	65-50882-3		TRANSMITTER AND SWITCH INSTL, ENG OIL PRESSURE AND LOW PRESSURE WARNING							C	RF
1	ST104Z		. TRANSMITTER, OIL PRESSURE, V61349							A	1
1	ST104AB		. TRANSMITTER, OIL PRESSURE, V61349							B	1
1A	418-20044		. TRANSMITTER, OIL PRESSURE, V14140							C	2
1B	BACN10DR12J		. NUT							C	2
2	MS21902-4		. UNION *[2] (PRE SB 79-1003)							AB	1
2A	66-13955-875		. VENT PORT *[1] (POST SB 79-1003)							AB	1
2A	66-13955-1		. VENT PORT (OPT) *[1]							AB	1
3	MS21916-6-4		. REDUCER UNION							AB	1
4	BACP11PB4		. PACKING *[2] (PRE SB 79-1003)							AB	2
4	BACP11PB4		. PACKING *[1] (POST SB 79-1003)							AB	1
4A	BACP11PB4A		. PACKING *[1]							AB	1
4A	BACG10U4		. PACKING *[1] (POST SB 79-1003)							AB	1
5	K710-22		. SHOCK MOUNT, V51116							AB	1
6	BACS12CB3-9		. SCREW							AB	4
7	AN960-10L		. WASHER							AB	4
8	BACN10JC3		. NUT (REPLS NAS679A3W)							AB	4
8	NAS679A3W		. NUT (REPLD BY BACN10JC3)							AB	4
9	BACS12CB06-4		. SCREW							AB	1
10	AN960-6L		. WASHER							AB	1
11	MS35338-41		. LOCKWASHER							AB	1
12	MS25083-2AB5		. BONDING JUMPER							AB	1
13	BACS12CB3-5		. SCREW							AB	1
14	AN960-10L		. WASHER							AB	1
15	MS35338-43		. LOCKWASHER							AB	1
16	BACN10JC3		. NUT (REPLS NAS679A3)							AB	1
16	NAS679A3		. NUT (REPLD BY BACN10JC3)							AB	1
17	42D107A1		. PRESSURE SWITCH, V09049 (BOEING 10-3269-8)								1
18	BACS12CB3-7		. SCREW							AB	2
18	NAS1801-3-7		. SCREW							C	2
19	AN960-10L		. WASHER								2
20	BACN10JC3		. NUT (REPLS NAS679A3W)								2
20	NAS679A3W		. NUT (REPLD BY BACN10JC3)							AB	2
21	66-19214-1		. FITTING, ELBOW								1
22	69-45458-1		. BOLT								1

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1115-23	BACP11PB4		.	P	A	C	K	I	N	G	AB	2
23	BACP11PB4		.	P	A	C	K	I	N	G	AB	1
23	BACP11PB4A		.	P	A	C	K	I	N	G	C	2
24	BACP11P11		.	P	A	C	K	I	N	G	AB	2
24	BACP11P11		.	P	A	C	K	I	N	G		1
25	NAS1236-4H		.	B	O	L	T	,	U	N	I	1
			.	B	O	L	T	,	U	N	I	1
			.	B	O	L	T	,	U	N	I	1
26	NAS1237-4		.	E	L	B	O	W	,	U	N	1
			.	E	L	B	O	W	,	U	N	1
			.	E	L	B	O	W	,	U	N	1
27	66-13955-875		.	V	E	N	T	P	O	R	T	1
27	66-13955-1		.	V	E	N	T	P	O	R	T	1
28	BACP11PB4A		.	P	A	C	K	I	N	G	AB	1
28	BACG10U4		.	P	A	C	K	I	N	G	AB	1

- *[1] LIMITED USAGE. USED ON TRANSMITTERS AND SWITCHES NOT VENTED TO ENGINE GEARBOX.
- *[2] LIMITED USAGE. USED ON TRANSMITTERS AND SWITCHES VENTED TO ENGINE GEARBOX.



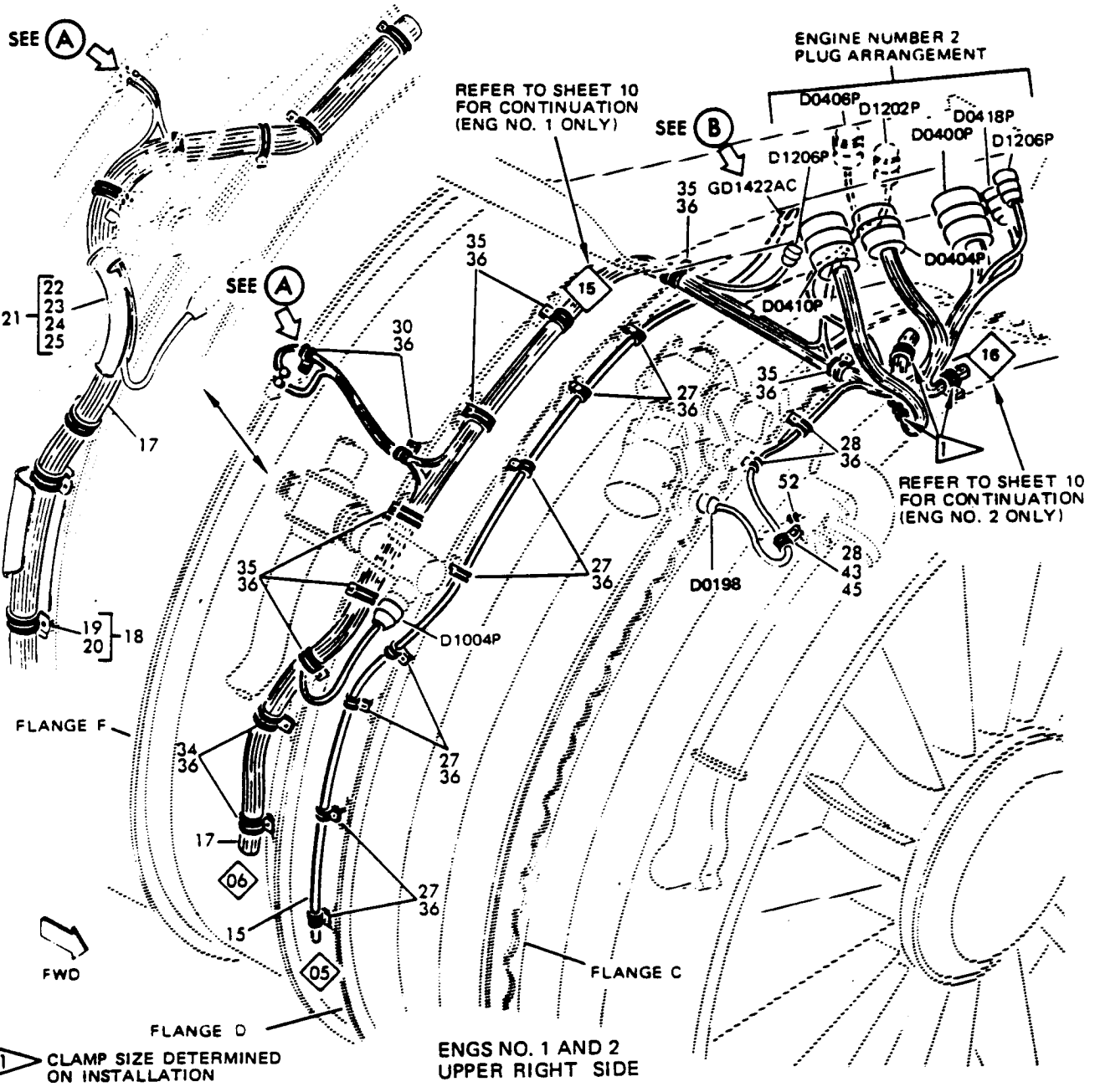
REF.
FIG
519
551

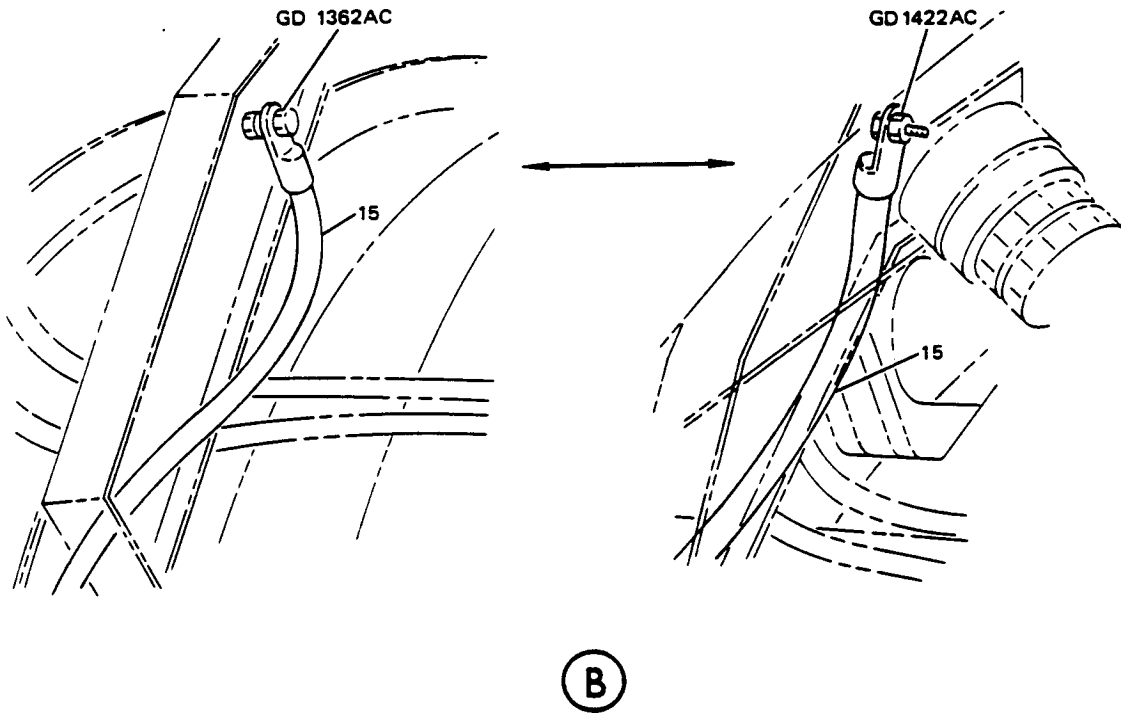
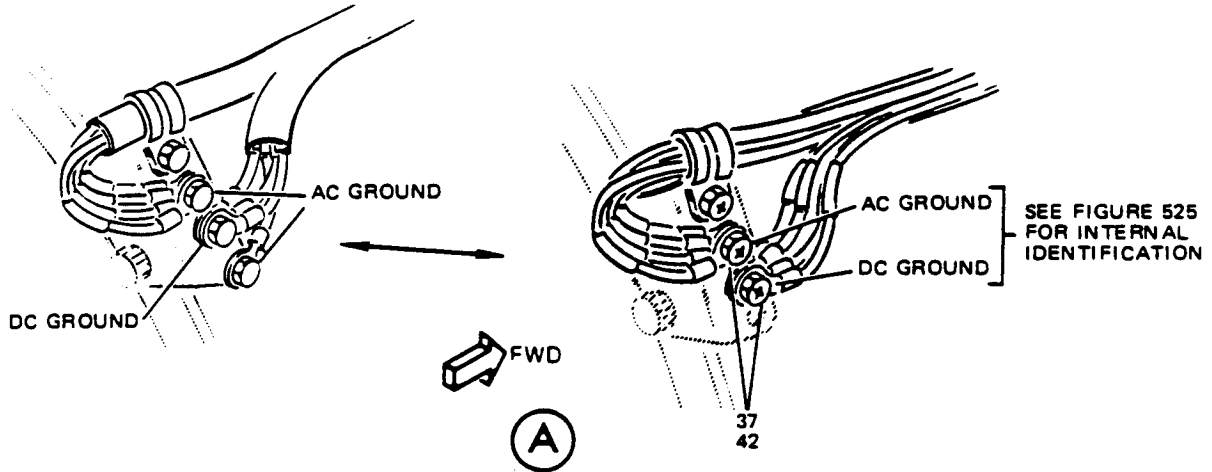
Engine Drain System Drain Fitting Installation
Figure 1116

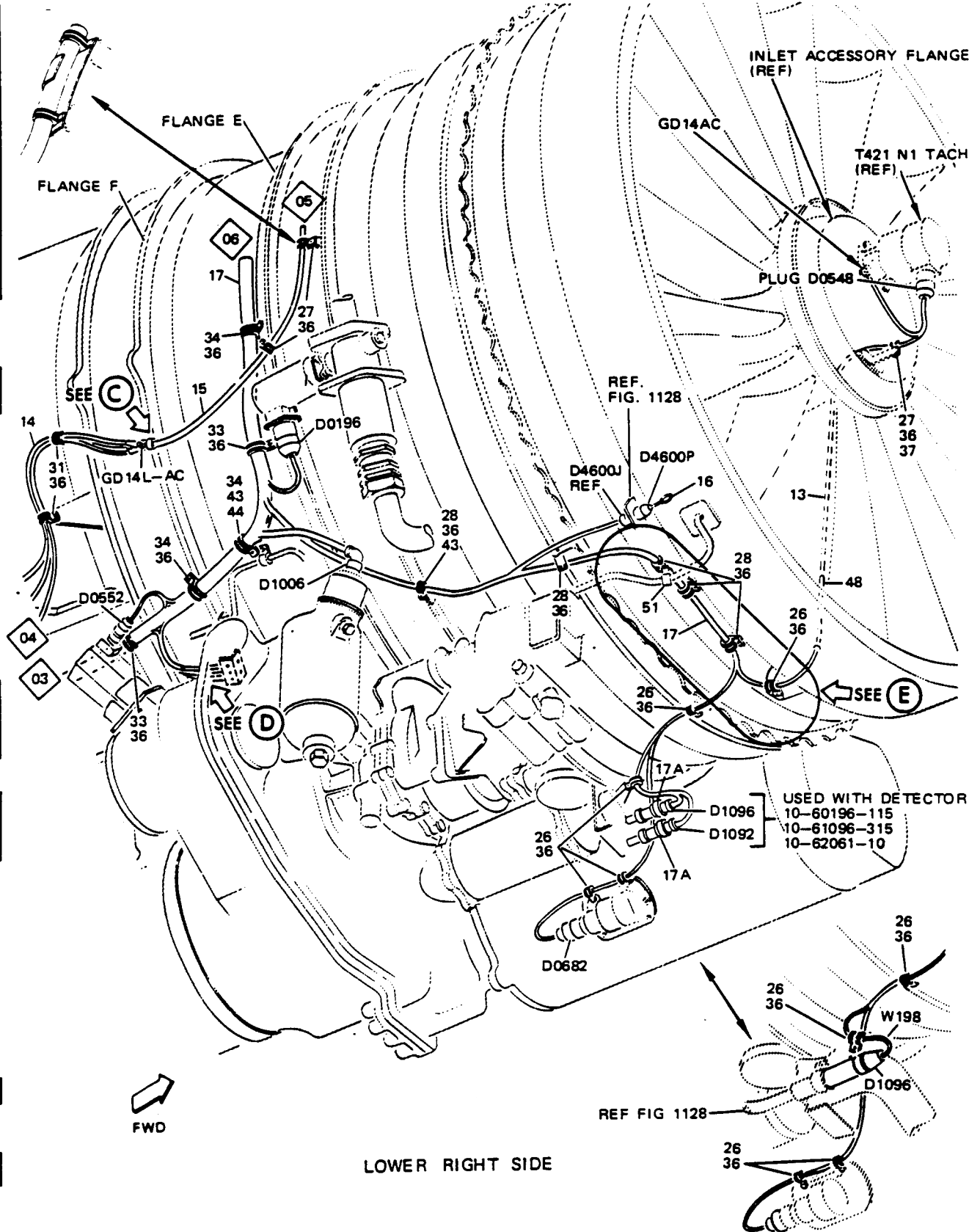
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1116-	65-52261-2		DRAIN FITTING INSTL, ENGINE DRAIN SYSTEM (BASIC)							A	RF
	65-52261-1		DRAIN FITTING INSTL, ENGINE DRAIN SYSTEM (VARIABLE)							B	RF
1	69-39494-1		. SEAL								1
2	69-38973-1		. MANIFOLD, DRAIN								1
3	BACS12CB3-6		. SCREW								3
4	AN960-10L		. WASHER								3
5	NAS679A3W		. NUT								3
6	69-39494-1		. SEAL								1
7	69-39487-1		. FITTING, DRAIN (PRE SB 71-1162)							B	1
7	69-62277-1		. FITTING, DRAIN							A	1
7	69-62277-1		. FITTING, DRAIN (POST SB 71-1055, 71-1058)							B	1
8	BACS12CB3-7		. SCREW								2
9	AN960-10L		. WASHER								2
10	NAS679A3W		. NUT								2
11	69-39494-2		. SEAL								1
12	69-52283-1		. BRACKET, FIRE DETECTOR *[1]							B	1
12	69-62249-2		. BRACKET, FIRE DETECTOR							A	1
12	69-62249-2		. BRACKET, FIRE DETECTOR *[2]							B	1
13	69-39486-4		. DRAIN FITTING								1
14	BACS12CB3-8		. SCREW								2
15	AN960-10L		. WASHER								2
16	NAS679A3W		. NUT								2

*[1] PRE SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1079, 71-1081, 71-1083, 71-1088, 71-1091, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151, 71-1162, 71-1179, 71-1189, 71-1198, 71-1278

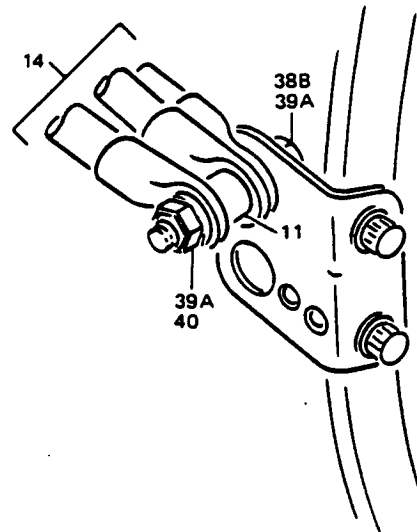
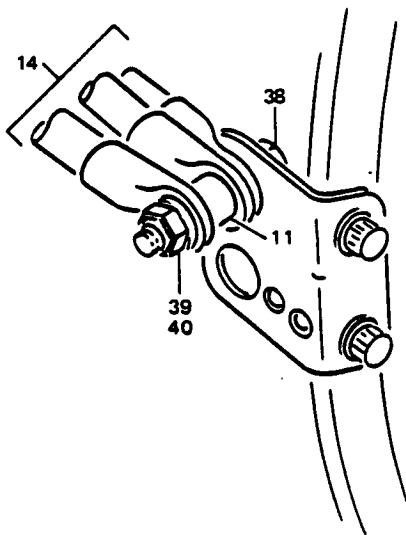
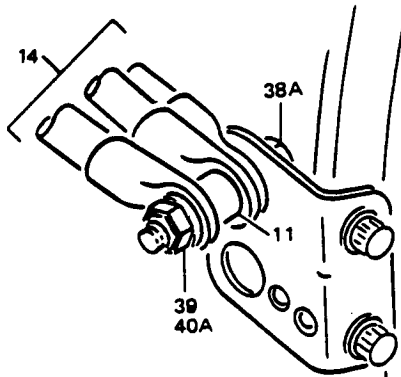
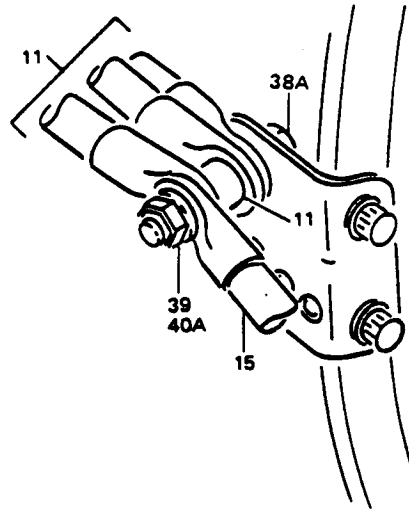
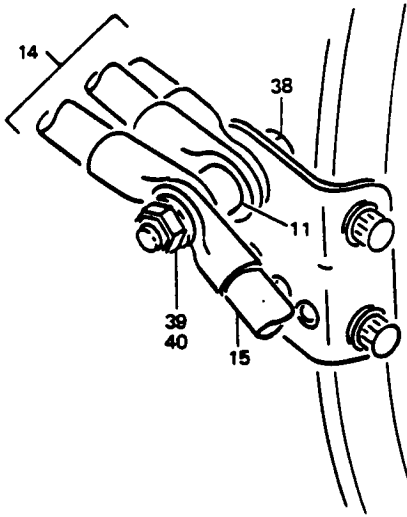
*[2] POST SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1079, 71-1081, 71-1083, 71-1088, 71-1091, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151, 71-1162, 71-1179, 71-1189, 71-1198, 71-1278



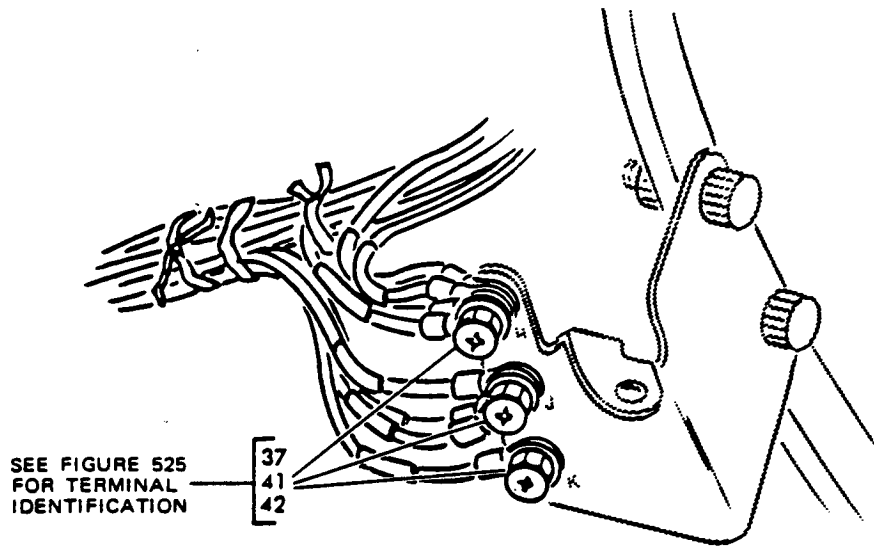




OVERHAUL MANUAL

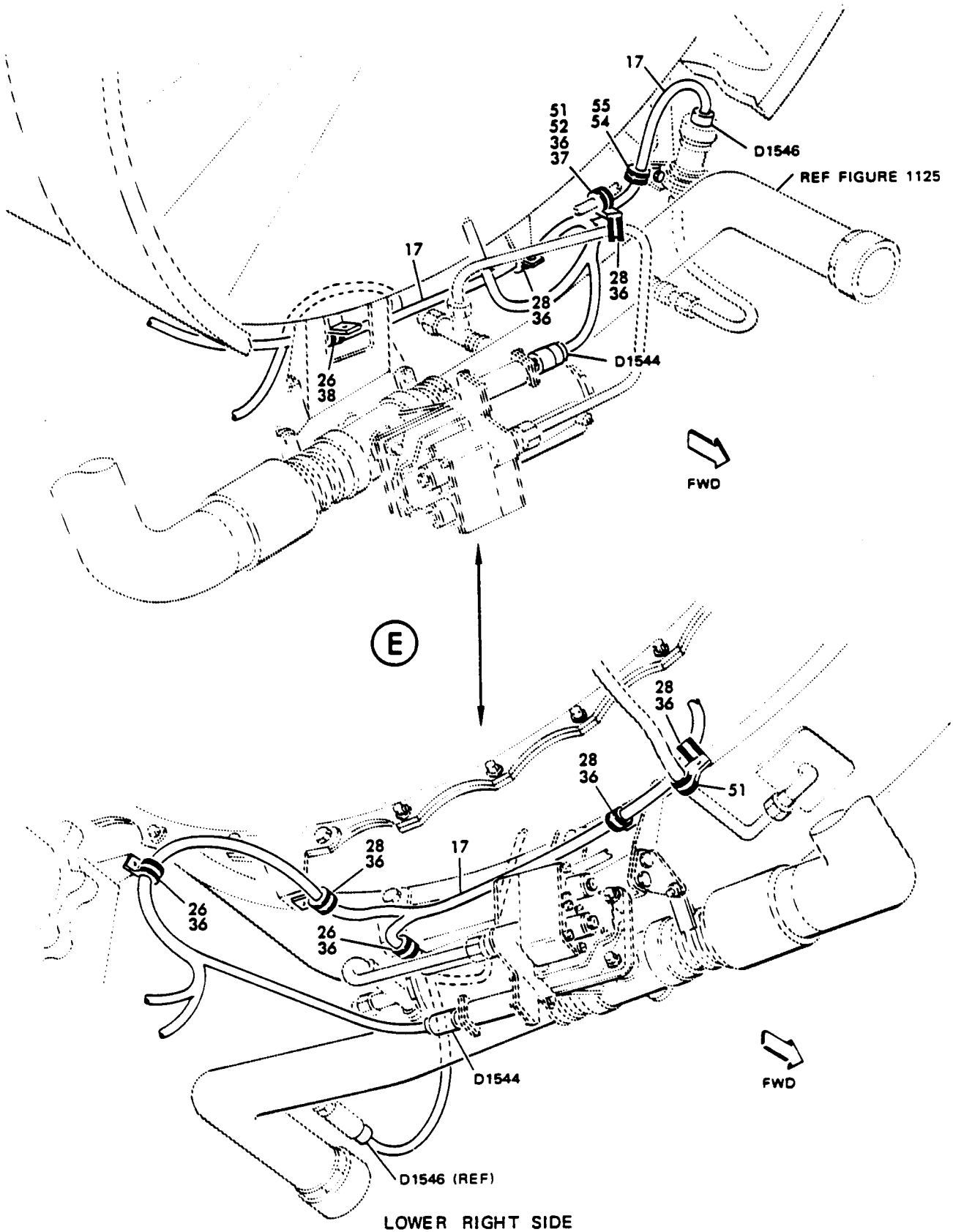


©

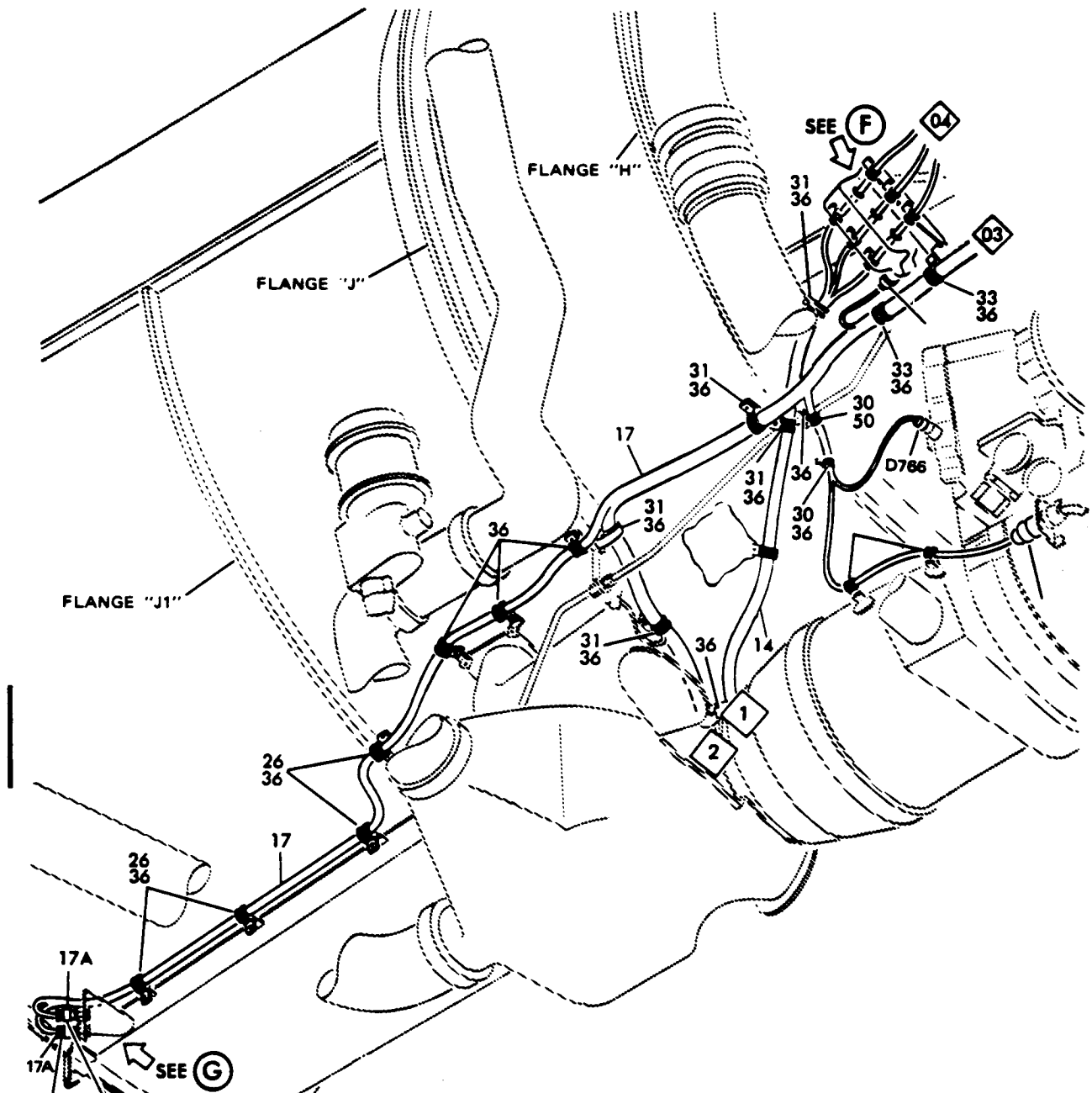


SEE FIGURE 525 FOR WIRE IDENTIFICATION

(D)

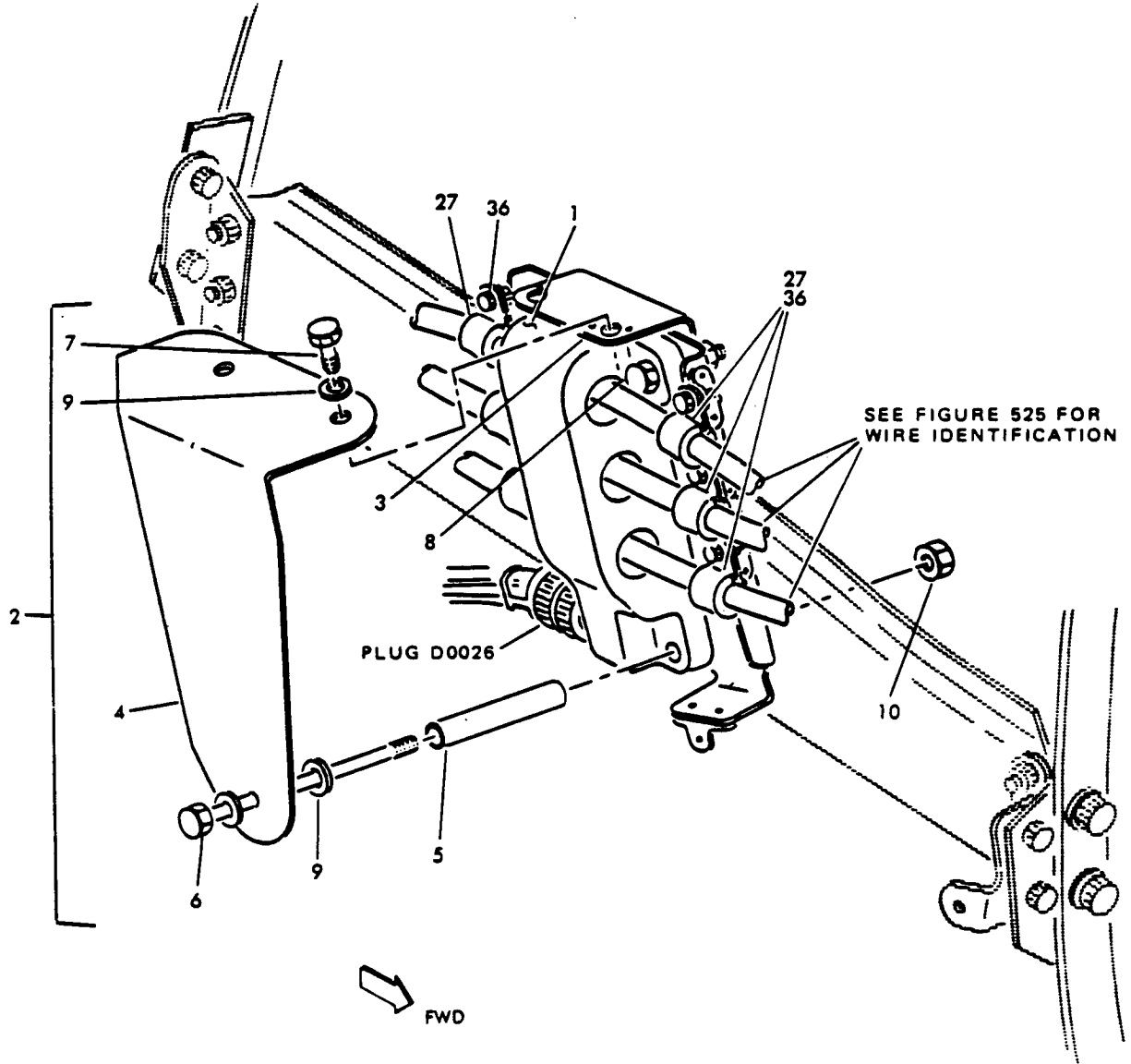


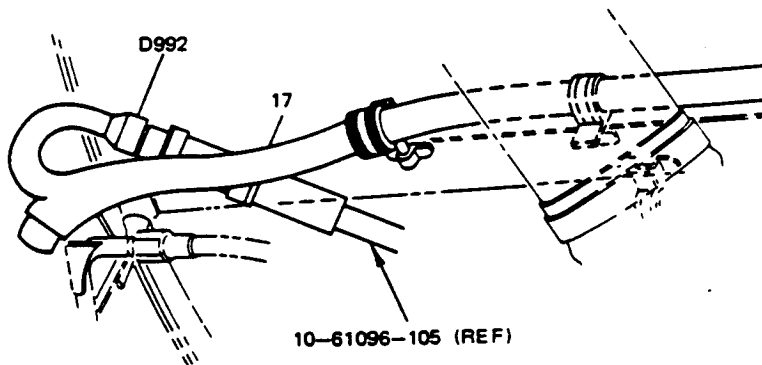
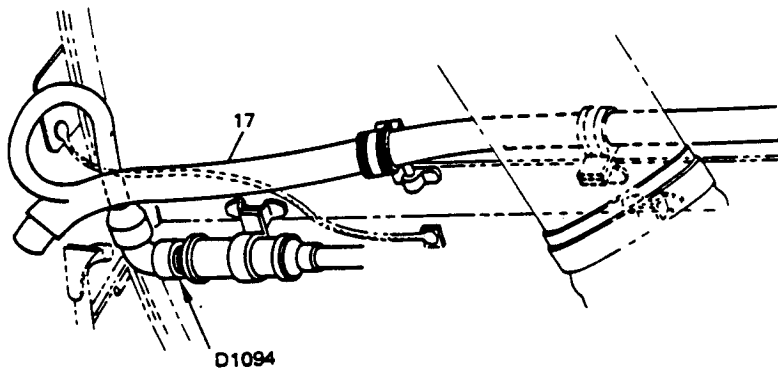
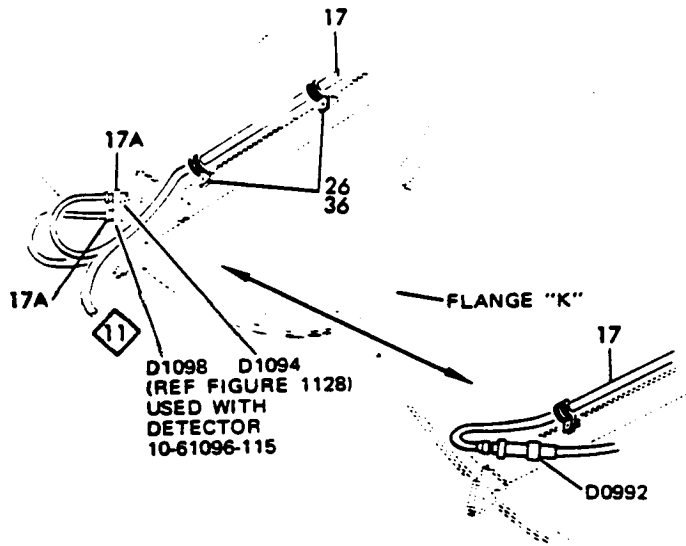
Electrical Installation
Figure 1117 (Sheet 6)



LOWER RIGHT SIDE

D1098 D1094
(REF FIGURE 1128)
USED WITH
DETECTOR
10-61096-115



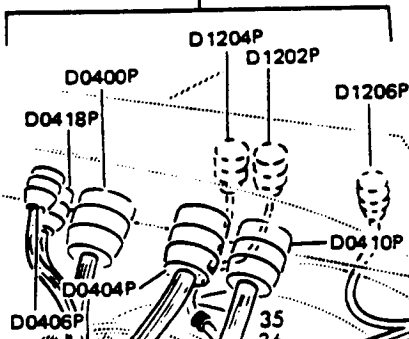


Electrical Installation
Figure 1117 (Sheet 9)

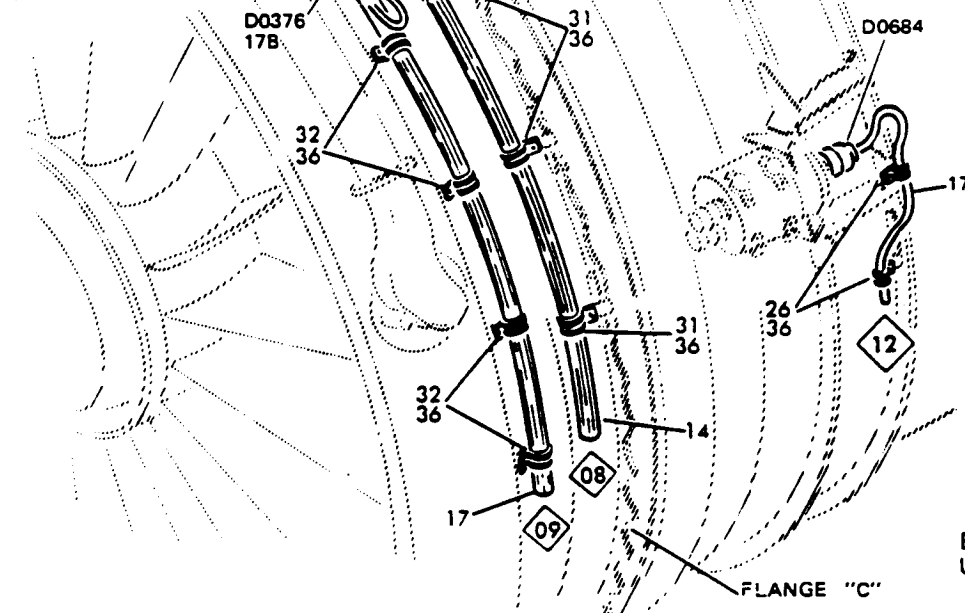
OVERHAUL MANUAL

REFER TO SHEET 1
FOR CONTINUATION
(ENG NO. 1 ONLY)

**ENGINE NUMBER 1
PLUG ARRANGEMENT**



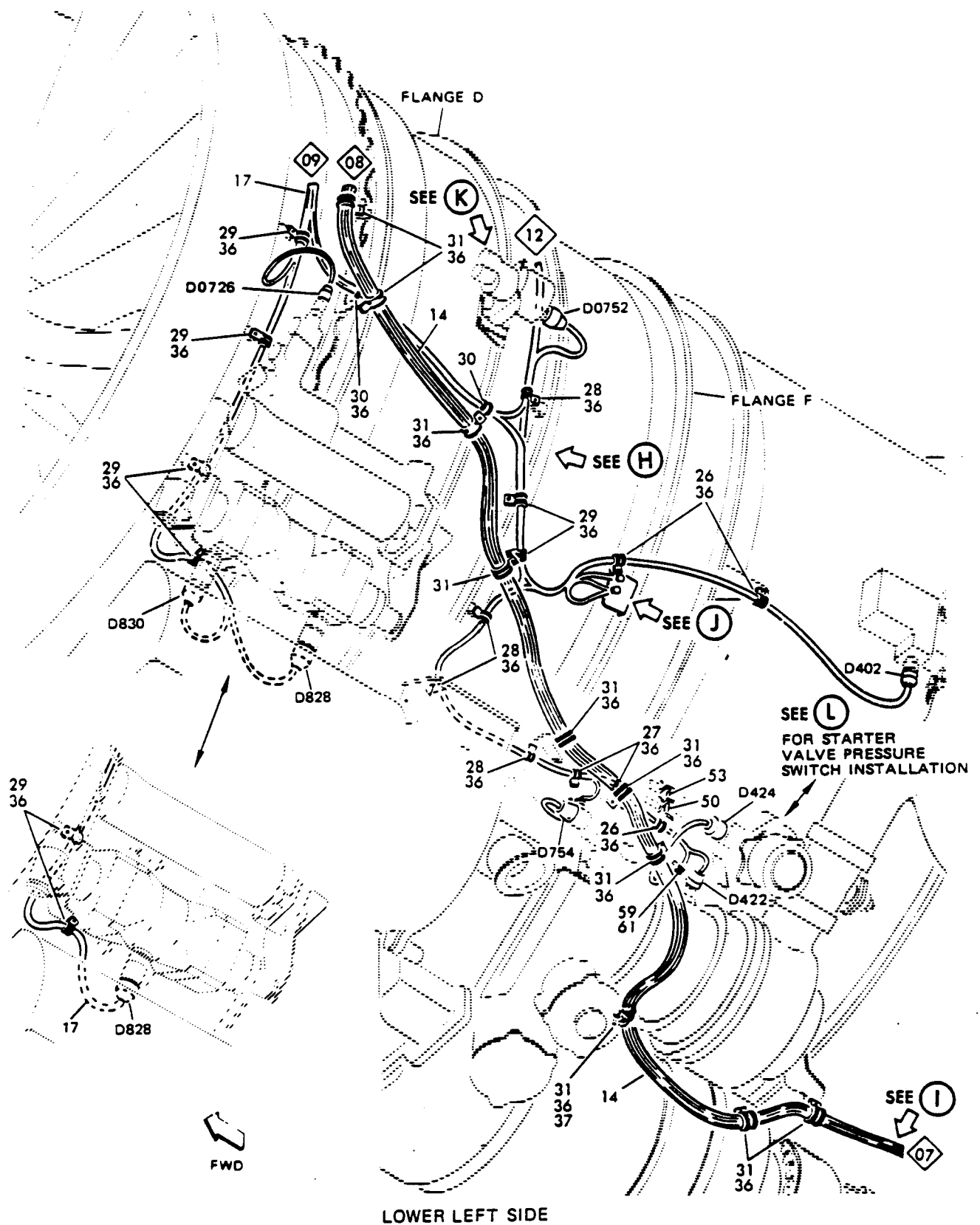
REFER TO SHEET 1 FOR
CONTINUATION
(ENG NO. 2 ONLY)



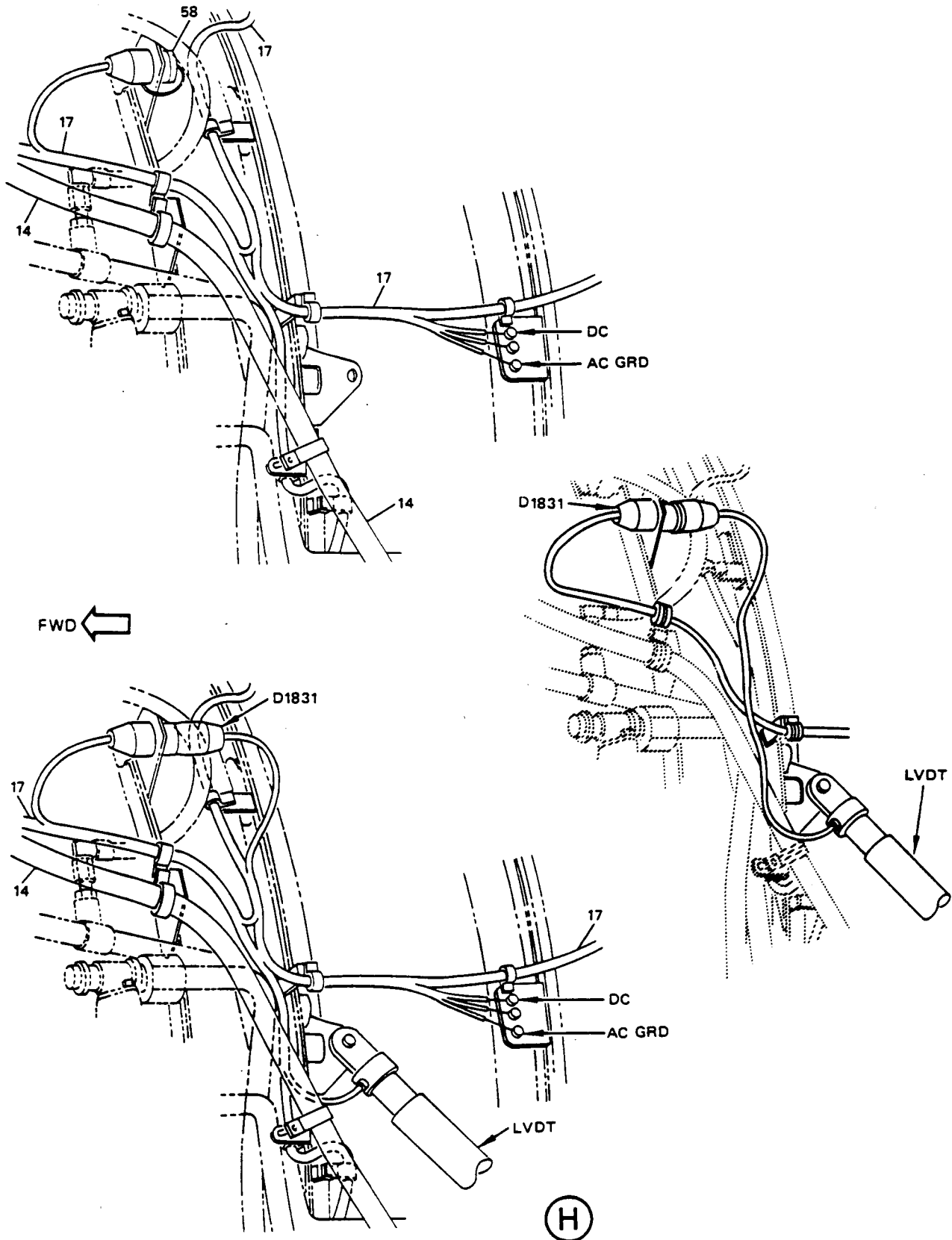
 CLAMP SIZE DETERMINED
ON ASSEMBLY

**ENGS NO. 1 AND 2
UPPER LEFT SIDE**

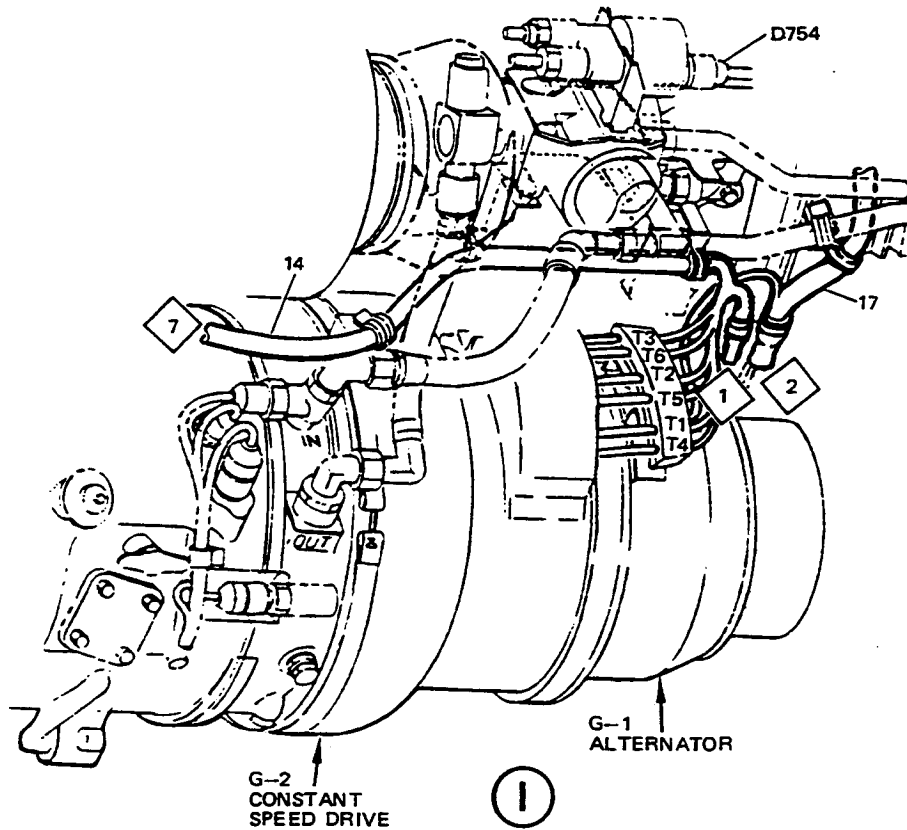
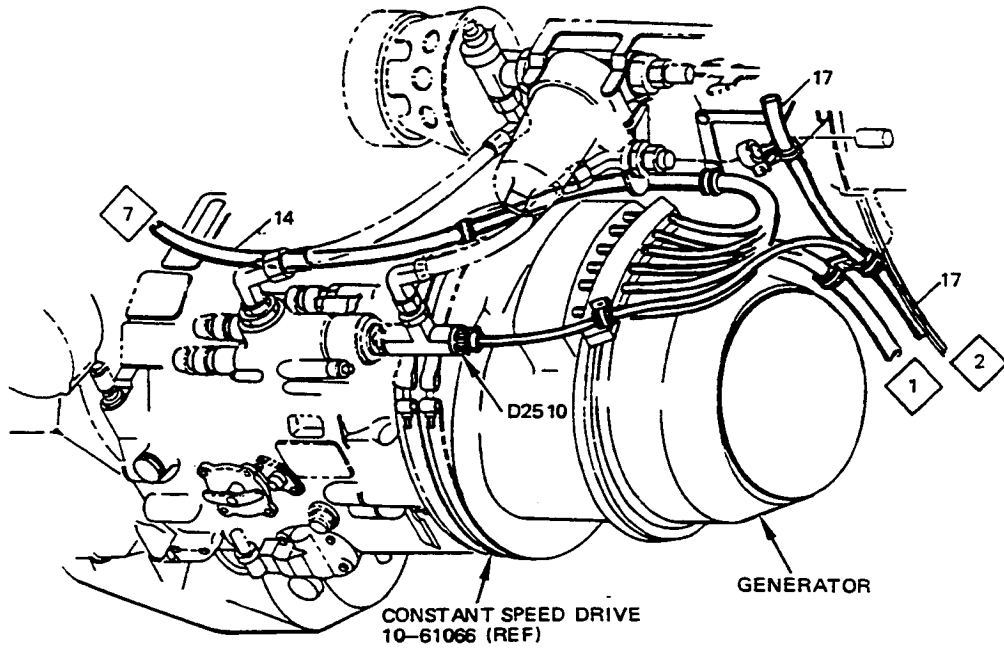
Electrical Installation
Figure 1117 (Sheet 10)



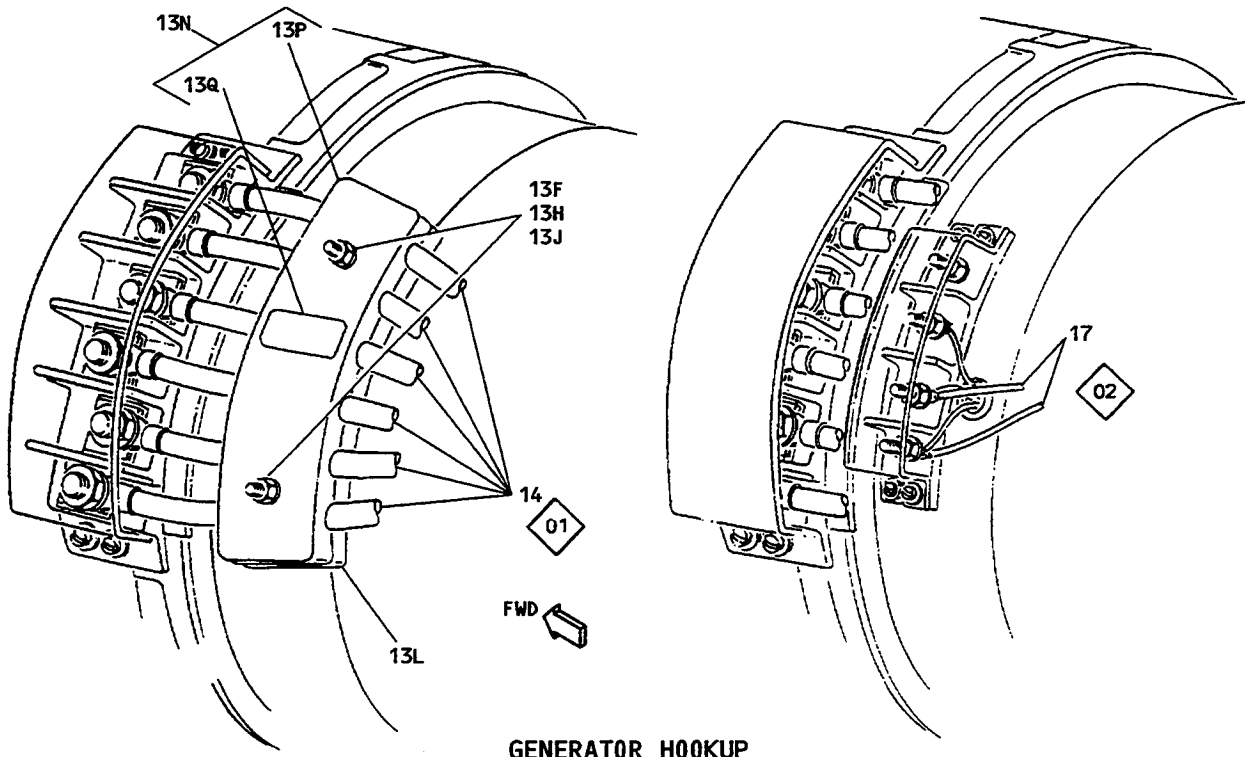
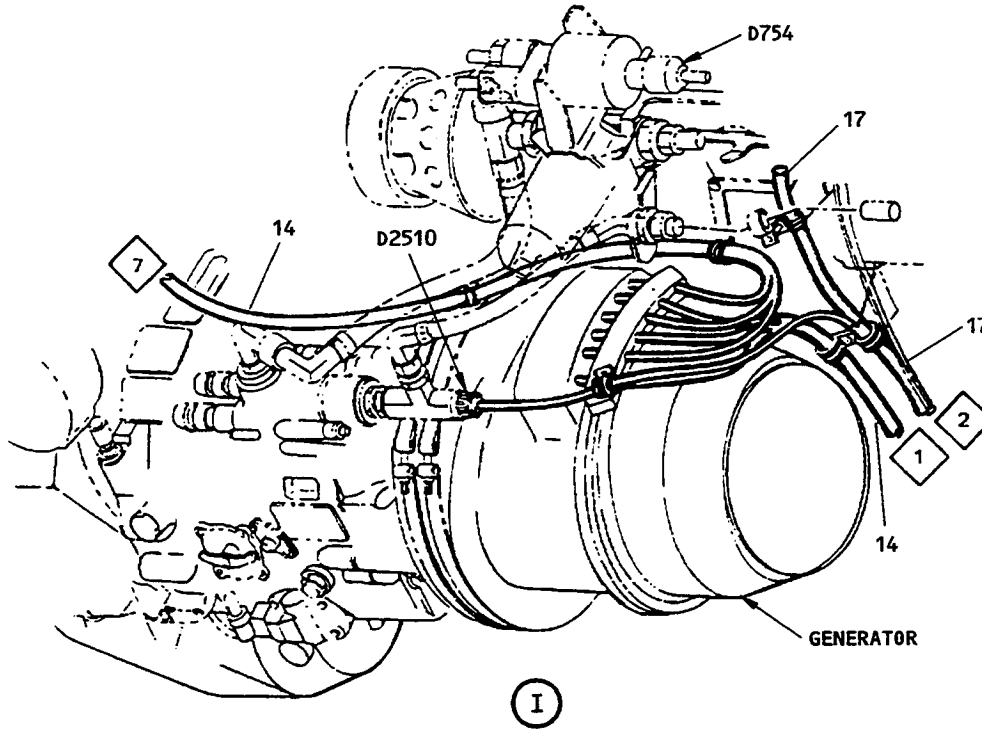
OVERHAUL MANUAL



Electrical Installation
Figure 1117 (Sheet 12)

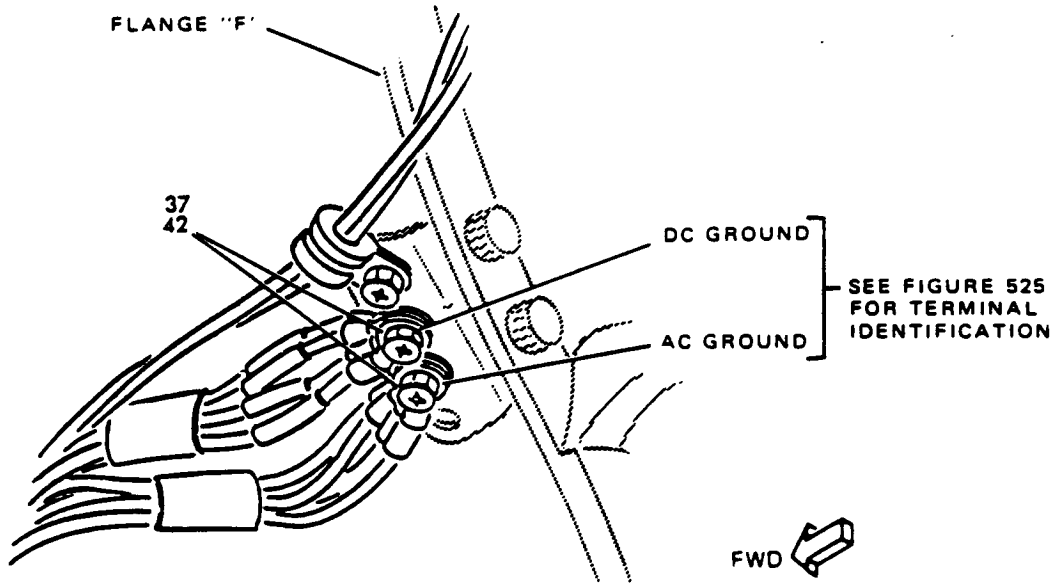


Electrical Installation
Figure 1117 (Sheet 13)

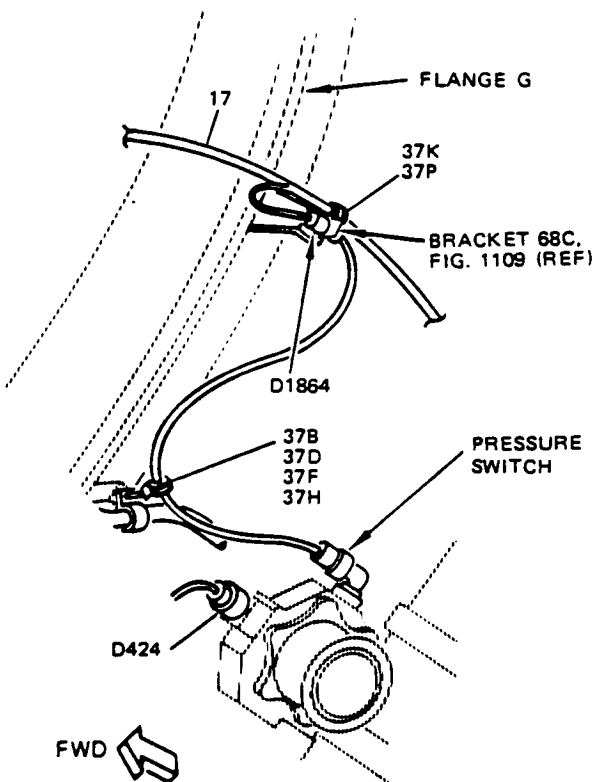


GENERATOR HOOKUP

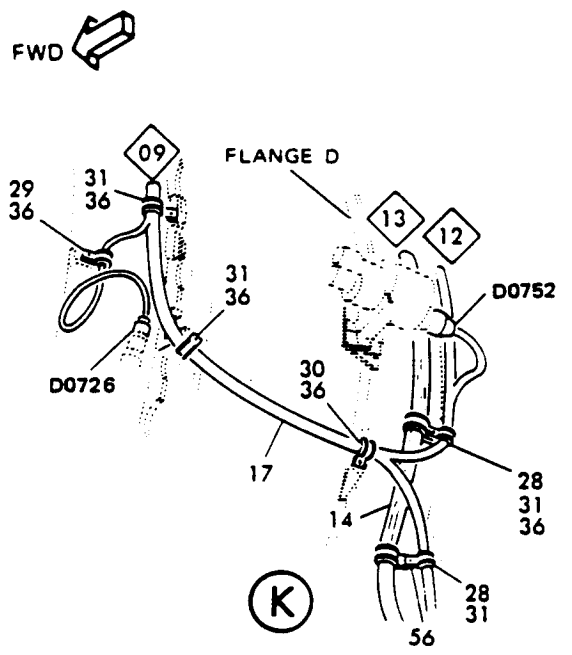
**Electrical Installation
Figure 1117 (Sheet 4)**



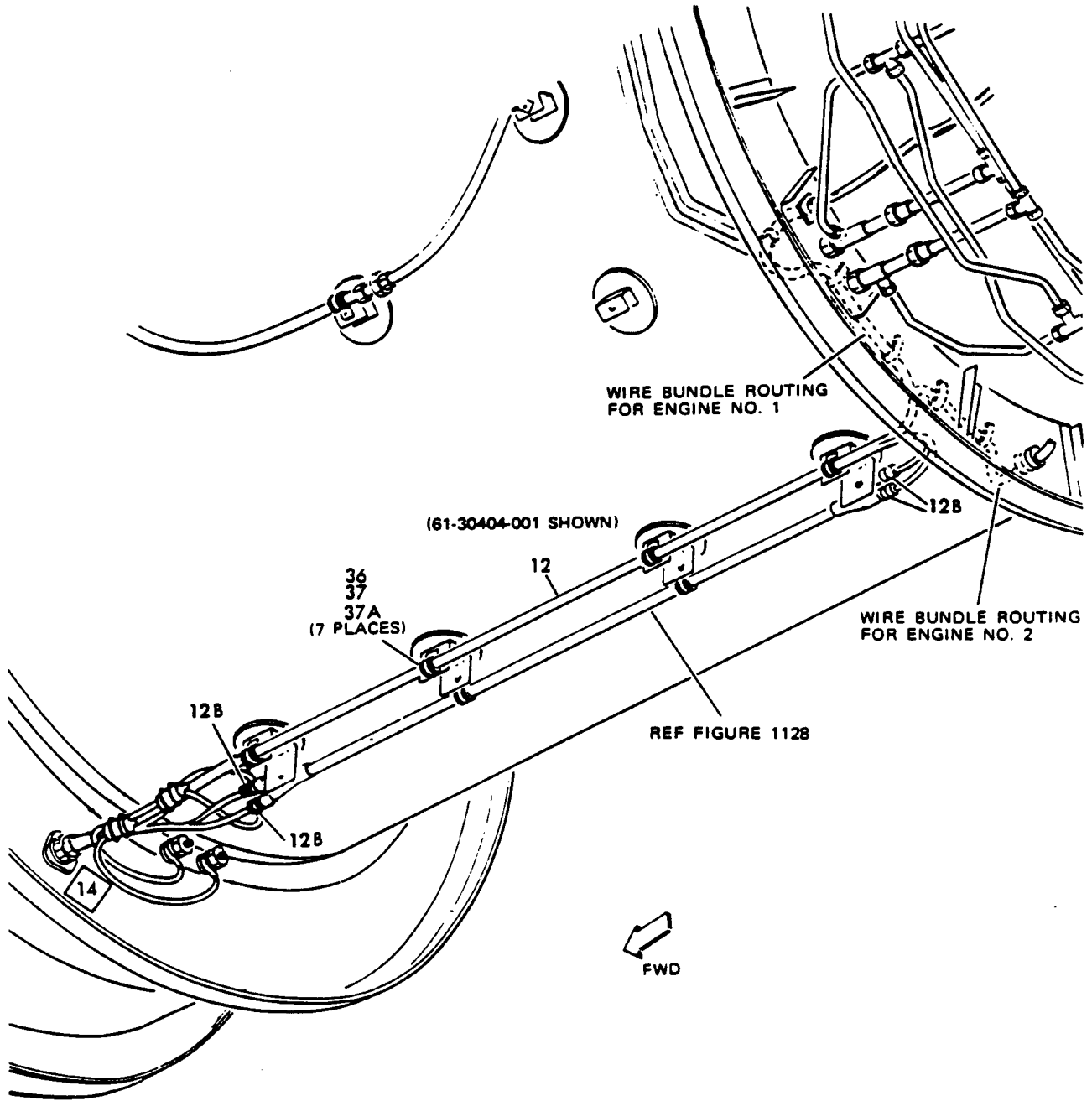
(J)



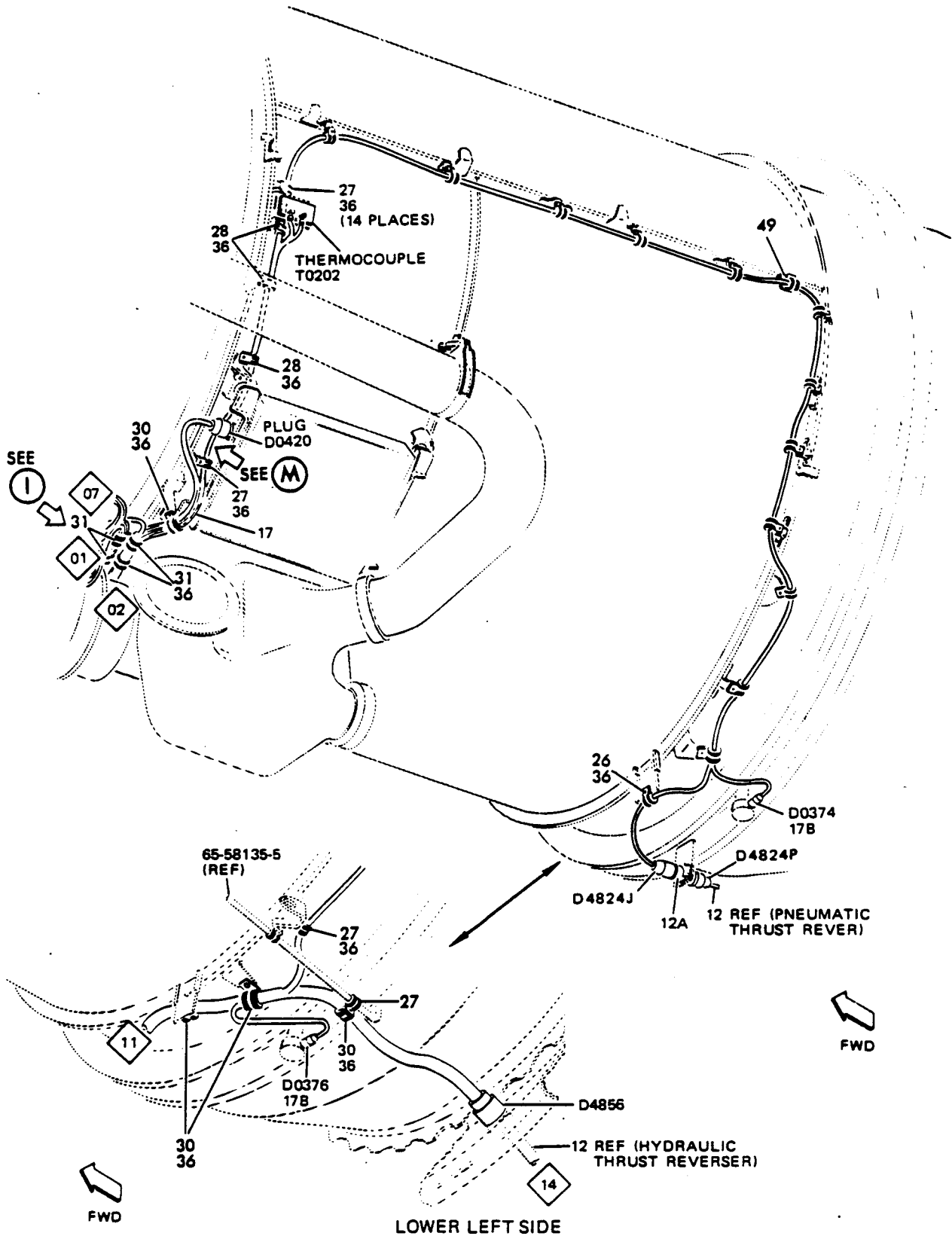
(L)



(K)

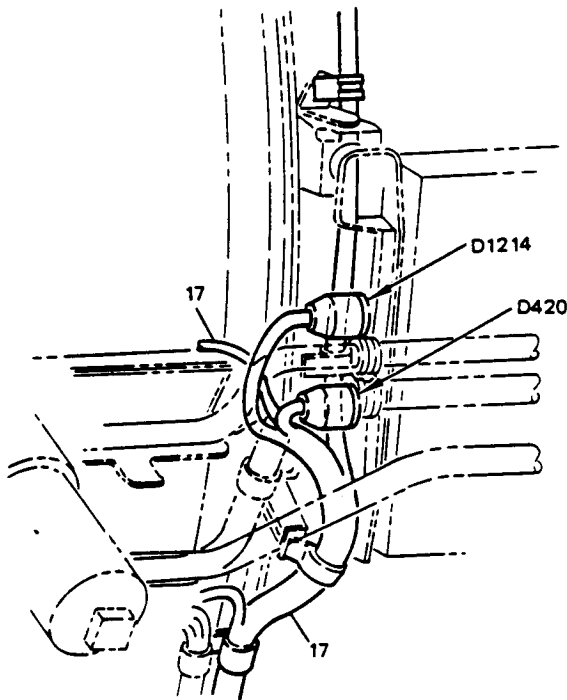
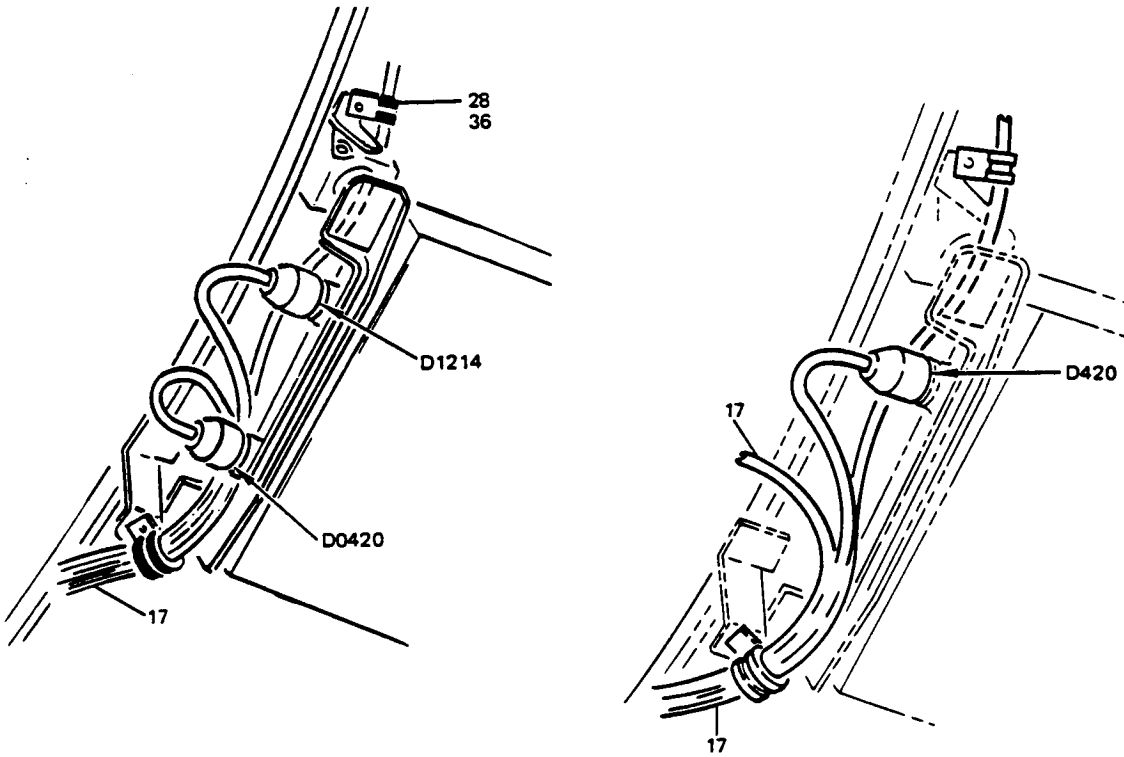


LOWER LEFT SIDE



Electrical Installation
Figure 1117 (Sheet 17)

OVERHAUL MANUAL



(M)

Electrical Installation
Figure 1117 (Sheet 18)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1117-	65-52971-1		ELECTRICAL	INSTL						A	RF	
	65-52971-5001		ELECTRICAL	INSTL (REPLD BY						B	RF	
		65-52971-5002		65-52971-5023) * [1]	ELECTRICAL	INSTL * [1]					C	RF
		65-52971-5003		ELECTRICAL	INSTL * [1]						D	RF
		65-52971-5004		ELECTRICAL	INSTL (REPLD BY						E	RF
				65-52971-5024) * [1]	ELECTRICAL	INSTL * [1]					F	RF
		65-52971-5005		ELECTRICAL	INSTL * [1]						G	RF
		65-52971-5006		ELECTRICAL	INSTL * [1]						H	RF
		65-52971-5007		ELECTRICAL	INSTL * [1]						I	RF
		65-52971-5008		ELECTRICAL	INSTL * [1]						J	RF
		65-52971-5009		ELECTRICAL	INSTL * [1]						K	RF
		65-52971-5010		ELECTRICAL	INSTL * [1]						L	RF
		65-52971-5011		ELECTRICAL	INSTL * [1]						M	RF
		65-52971-5012		ELECTRICAL	INSTL * [1]						N	RF
		65-52971-5013		ELECTRICAL	INSTL (REPLD BY						O	RF
				65-52971-5024) * [1]	ELECTRICAL	INSTL * [1]					P	RF
		65-52971-5014		ELECTRICAL	INSTL * [1]						Z	RF
		65-52971-5015		ELECTRICAL	INSTL * [1]						Q	RF
		65-52971-5016		ELECTRICAL	INSTL * [1]						R	RF
		65-52971-5017		ELECTRICAL	INSTL * [1]						S	RF
		65-52971-5018		ELECTRICAL	INSTL * [1]						T	RF
		65-52971-5019		ELECTRICAL	INSTL * [1]						U	RF
		65-52971-5020		ELECTRICAL	INSTL * [1]						V	RF
		65-52971-5021		ELECTRICAL	INSTL * [1]						W	RF
		65-52971-5022		ELECTRICAL	INSTL * [1]						X	RF
		65-52971-5023		ELECTRICAL	INSTL REPLS						Y	RF
			65-52971-5001) * [1]	ELECTRICAL	INSTL REPLS							
	65-52971-5024		65-52971-5004, -5013) * [1]	ELECTRICAL	INSTL * [1]							
1	941D339-1		. TRANSFORMER, CURRENT, V83843 (BOEING 10-61224-5)								1	
2	65-65358-1		. SHIELD INSTL, HEAT								1	
3	69-52860-1		. . BRACKET ASSY								1	
4	65-65358-2		. . SHIELD, HEAT								1	
5	NAS43DD3-128		. . SPACER								1	
6	BACS12CB3-48		. . SCREW								1	
7	BACS12CB3-7		. . SCREW								2	
8	BACS12CB3-14		. . SCREW								2	
9	AN960D10L		. . WASHER								6	
10	NAS679A3W		. . NUT								3	
11	66-18526-1		. SPACER, ELECTRICAL								1	

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-12	61-30404-001		. WIRE BUNDLE ASSY, FIRE DETECTOR (REPLD BY 61-30404-3)							X	1
12	61-30404-002		DELETED								
12	61-30404-1		. WIRE BUNDLE ASSY, FIRE DETECTOR							B D-Y	1
12	61-30404-2		. WIRE BUNDLE ASSY, FIRE DETECTOR							CEN	1
12	61-30404-3		. WIRE BUNDLE ASSY, FIRE DETECTOR (PREF)							EHN OXY	1
12	61-30404-011		. WIRE BUNDLE ASSY, FIRE DETECTOR*[4]							EHNOX	1
12	61-30404-003		. WIRE BUNDLE ASSY							B-E G-OQ	1
12A	BACS12CB3-5		. SCREW								4
12B	209592		. GASKET, V33525 *[3]								4
13	61-30196-001		. WIRE BUNDLE ASSY, TACHOMETER *[4]								1
13	61-30196-003		. WIRE BUNDLE ASSY, TACHOMETER *[4]							EHNOX	
13	61-30196-005		. WIRE BUNDLE ASSY, TACHOMETER *[4]							N	1
13F	NAS679A3W		. NUT, SELF-LOCKING								2
13H	AN960-10L		. WASHER								2
13J	NAS514P-1032-16		. SCREW								2
13L	69-40075-2		. SPACER, BOTTOM								1
13N	69-40075-4		. SPACER, TOP ASSY								1
13P	69-40075-1		. . SPACER, TOP								1
13Q	BACM10T19LS		. . MARKER, ALUMINUM FOIL								1
14	61-30200-001		. WIRE BUNDLE ASSY, AC GENERATOR								1
14	61-30200-003		. WIRE BUNDLE ASSY, AC GENERATOR*[4]							EN	1
14	61-30200-005		. WIRE BUNDLE ASSY, AC GENERATOR*[4]							EHNOX	1
-14	61-30200-006		. WIRE BUNDLE ASSY, AC GENERATOR*[4]							NE	1
14	61-30200-007		. WIRE BUNDLE ASSY, AC GENERATOR*[4]							N	1
15	61-30202-001		. WIRE BUNDLE ASSY, GROUND							B	1
16	61-30204-002		. WIRE BUNDLE ASSY, FIRE DETECTOR							C	1
16	61-30204-001		. WIRE BUNDLE ASSY, FIRE DETECTOR							C	1
16	61-30204-2		. WIRE BUNDLE ASSY, FIRE DETECTOR							EN	1
16	61-30204-3		. WIRE BUNDLE ASSY, FIRE DETECTOR (PREF)							EN	1
17	61-30198-005		. WIRE BUNDLE ASSY, MAIN (REPLD BY 61-30198-064) *[4]							F	1
17	61-30198-006		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-007		. WIRE BUNDLE ASSY, MAIN *[4]							H	1
17	61-30198-009		. WIRE BUNDLE ASSY, MAIN *[4]							IQ	1
17	61-30198-011		. WIRE BUNDLE ASSY, MAIN *[4]							L	1
17	61-30198-012		. WIRE BUNDLE ASSY, MAIN *[4]							K	1
17	61-30198-015		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-016		. WIRE BUNDLE ASSY, MAIN *[4]							O	1
17	61-30198-026		. WIRE BUNDLE ASSY, MAIN *[4]							D	1

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY															
			1	2	3	4	5	6	7																	
1117-17	61-30198-027		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	*[4]	H	1		
17	61-30198-028		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-029		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-030		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-031		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-032		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	*[4]	D	1		
17	61-30198-033		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-034		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-035																									
17	61-30198-036		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-037		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-038		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-039		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-040		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-044		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	*[4]	GJ	1		
17	61-30198-045		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-047		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	*[4]	P	1		
17	61-30198-054																									
17	61-30198-056		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	*[4]	M	1		
17	61-30198-057		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-060																									
17	61-30198-061																									
17	61-30198-062		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-064		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-065		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-066		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-067		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	
17	61-30198-068		.	W	I	R	E	B	U	N	D	L	E	A	S	S	Y,	M	A	I	N	(R	E	P	L	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY								
			1	2	3	4	5	6	7										
1117-17	61-30198-069		.	W	I	R	E	B	U	N	D	L	E	A	S	S		D	1
17	61-30198-070		.	W	I	R	E	B	U	N	D	L	E	A	S	S		EIK	1
17	61-30198-071		.	W	I	R	E	B	U	N	D	L	E	A	S	S		QRU	1
17	61-30198-072		.	W	I	R	E	B	U	N	D	L	E	A	S	S		NW	1
17	61-30198-073		.	W	I	R	E	B	U	N	D	L	E	A	S	S		H	1
17	61-30198-074		.	W	I	R	E	B	U	N	D	L	E	A	S	S		M	1
17	61-30198-075		.	W	I	R	E	B	U	N	D	L	E	A	S	S		B	1
17	61-30198-076		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-077		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-079		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-080		.	W	I	R	E	B	U	N	D	L	E	A	S	S		F	1
17	61-30198-081		.	W	I	R	E	B	U	N	D	L	E	A	S	S		H	1
17	61-30198-082		.	W	I	R	E	B	U	N	D	L	E	A	S	S		EIKNO	1
17	61-30198-083		.	W	I	R	E	B	U	N	D	L	E	A	S	S		PQRUV	1
17	61-30198-084		.	W	I	R	E	B	U	N	D	L	E	A	S	S		B	1
17	61-30198-085		.	W	I	R	E	B	U	N	D	L	E	A	S	S		B	1
17	61-30198-088		.	W	I	R	E	B	U	N	D	L	E	A	S	S		M	1
17	61-30198-089		.	W	I	R	E	B	U	N	D	L	E	A	S	S		NW	1
17	61-30198-090		.	W	I	R	E	B	U	N	D	L	E	A	S	S		X	1
17	61-30198-091		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-091		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-091		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY								
			1	2	3	4	5	6	7										
1117-17	61-30198-092		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-093		.	W	I	R	E	B	U	N	D	L	E	A	S	S		Y	1
17	61-30198-097		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-099		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-100		.	W	I	R	E	B	U	N	D	L	E	A	S	S		B	1
17	61-30198-101		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-105		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-107		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-110		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-111		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-112		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-113		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-114		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-115		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1
17	61-30198-116		.	W	I	R	E	B	U	N	D	L	E	A	S	S		E	1
17	61-30198-117		.	W	I	R	E	B	U	N	D	L	E	A	S	S		N	1

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY									
			1	2	3	4	5	6	7											
1117-17	61-30198-118		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-119		.	W	I	R	E	B	U	N	D	L	E	A	S	S			H	1
17	61-30198-120		.	W	I	R	E	B	U	N	D	L	E	A	S	S			EO	1
17	61-30198-121		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-122		.	W	I	R	E	B	U	N	D	L	E	A	S	S			X	1
17	61-30198-123		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-124		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-125		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-126		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-127		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-128		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-129		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-130		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-131		.	W	I	R	E	B	U	N	D	L	E	A	S	S			EN	1
17	61-30198-132																			
17	61-30198-133		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-134		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY									
			1	2	3	4	5	6	7											
1117-17	61-30198-135		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-136		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-137		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-138		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-140		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-141		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-142		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-143		.	W	I	R	E	B	U	N	D	L	E	A	S	S			EN	1
17	61-30198-144		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-145		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-146		.	W	I	R	E	B	U	N	D	L	E	A	S	S			E	1
17	61-30198-147		.	W	I	R	E	B	U	N	D	L	E	A	S	S			Y	1
17	61-30198-148		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-149		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-150		.	W	I	R	E	B	U	N	D	L	E	A	S	S			O	1
17	61-30198-151		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1
17	61-30198-152		.	W	I	R	E	B	U	N	D	L	E	A	S	S			N	1

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY																																																				
			1	2	3	4	5	6	7																																																						
1117-17	61-30198-153		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	2	4)*	[4]	N		1																			
17	61-30198-154		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	2	8)*	[4]	N		1																			
17	61-30198-155		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	0)*	[4]	E		1																			
17	61-30198-156		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	1	0)*	[4]	E		1																			
17	61-30198-157		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	3)*	[4]	N		1																			
17	61-30198-158		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	4)*	[4]	N		1																			
17	61-30198-159		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	5)*	[4]	N		1																			
17	61-30198-160		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	6)*	[4]	E		1																			
17	61-30198-161		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	7)*	[4]	N		1																			
17	61-30198-162		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	3	8)*	[4]	E		1																			
17	61-30198-163		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	0)*	[4]	N		1																			
17	61-30198-164		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	1)*	[4]	E		1																			
17	61-30198-165		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	2)*	[4]	N		1																			
17	61-30198-166		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	7	0)(R	E	P	L	D	B	Y	6	1	-	3	0	1	9	8	-	1	7	7)*	[4]	E	N	1
17	61-30198-167		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	4)*	[4]	E		1																			
17	61-30198-168		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	5)*	[4]	E		1																			
17	61-30198-169		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	4	6)*	[4]	E		1																			
17	61-30198-170		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	1	7)(R	E	P	L	D	B	Y	6	1	-	3	0	1	9	8	-	1	7	1)*	[4]	N		1
17	61-30198-171		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	7	0)*	[4]	N		1																			
17	61-30198-172		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	S	6	1	-	3	0	1	9	8	-	1	2	8)*	[4]	N		1																			

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-17	61-30198-211		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-131, 170) (REPLD BY 61-30198-181) *[4]							EN	1
17	61-30198-174		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-143) (REPLD BY 61-30198-178) *[4]							N	1
17	61-30198-175		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-170) *[4]							N	1
17	61-30198-176		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-070) *[4]							V	1
17	61-30198-177		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-166) *[4]							N	1
17	61-30198-178		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-174) *[4]							N	1
17	61-30198-179		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-116) *[4]							E	1
17	61-30198-181		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-173) *[4]							N	1
17	61-30198-185		. WIRE BUNDLE ASSY, MAIN (REPLS 61-30198-170) *[4]							N	1
17	61-30198-191		. WIRE BUNDLE ASSY, MAIN							N	1
17	61-30198-193		. WIRE BUNDLE ASSY, MAIN							E	1
17	61-30198-195		. WIRE BUNDLE ASSY, MAIN							E	1
17	61-30198-200		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-201		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-202		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-204		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-205		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-206		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-207		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-208		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-209		. WIRE BUNDLE ASSY, MAIN *[4]							X	1
17	61-30198-211		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-213		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-214		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-215		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-216		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-217		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-218		. WIRE BUNDLE ASSY, MAIN *[4]							N	1
17	61-30198-222		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-223		. WIRE BUNDLE ASSY, MAIN *[4]							E	1
17	61-30198-224		. WIRE BUNDLE ASSY, MAIN *[4]							N	1

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-											
17	61-30198-226		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-227		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-228		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-229		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-230		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-231		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		H	1
17	61-30198-232		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-233		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-234		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-235		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-236		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-237		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-238		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-239		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-240		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-241		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-242		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-243		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-244		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-245		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-246		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-247		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-248		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-249		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-250		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-251		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-252		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-253		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-254		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-255		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-256		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-257		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		X	1
17	61-30198-258		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-259		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-260		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-261		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-262		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-263		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-264		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-267		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-268		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-269		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-270		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-271		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY																				
			1	2	3	4	5	6	7																						
1117-																															
17	61-30198-273		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-274		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-275		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-276		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-277		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-278		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-280		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-281		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-282		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-283		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-284		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-285		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-286		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-289		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-290		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-291		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	D		B	Y	N	1
				61-30198-364)	*	[4]																							
17	61-30198-292		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-293		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-294		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-295		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	P	L	D		B	Y	N	1
				61-30198-333)	*	[4]																							
17	61-30198-296		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-297		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-298		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-299		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-300		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-301		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	H	1					
17	61-30198-302		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-303		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-304		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-305		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-306		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-307		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-308		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-309		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-310		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-311		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-312		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					
17	61-30198-313		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-314		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-315		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-316		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	N	1					
17	61-30198-317		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*	[4]	E	1					

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-											
17	61-30198-318		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-319		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-320		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-321		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-322		.	WIRE	BUNDLE	ASSY,	MAIN	(REPLD BY 61-30198-335) *[4]		N	1
17	61-30198-323		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-324		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-325		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-326		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-327		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-328		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-329		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-330		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-331		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-332		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-333		.	WIRE	BUNDLE	ASSY,	MAIN	(REPLS 61-30198-335) *[4]		N	1
-17	61-30198-335		.	WIRE	BUNDLE	ASSY,	MAIN	(REPLS 61-30198-322) *[4]		N	1
17	61-30198-336		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-337		.	WIRE	BUNDLE	ASSY,	MAIN	(REPLS 61-30198-291) *[4]		N	1
17	61-30198-338		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		H	1
17	61-30198-339		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-340		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-341		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-342		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-343		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-344		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		H	1
17	61-30198-345		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-346		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-347		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-348		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		EN	1
17	61-30198-349		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-350		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-351		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-352		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-353		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-354		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-355		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-356		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		E	1
17	61-30198-357		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1
17	61-30198-358		.	WIRE	BUNDLE	ASSY,	MAIN	*[4]		N	1

-ITEM NOT ILLUSTRATED

737

POWERPLANT BUILDUP

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY												
			1	2	3	4	5	6	7														
1117-																							
17	61-30198-359		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-360		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-361		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	H	1
17	61-30198-364		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	(R	E	1
17	61-30198-366		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-368		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-369		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-370		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-371		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-372		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-373		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-374		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-375		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-380		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-381		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-383		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-395		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-402		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-403		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-404		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-409		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-412		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-413		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-417		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-424		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-425		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	O	1
17	61-30198-426		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-427		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-428		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-429		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-430		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-431		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-432		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-433		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-434		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-435		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-436		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-437		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	N	1
17	61-30198-438		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-439		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	X	1
17	61-30198-440		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1
17	61-30198-441		.	W	I	R	E	B	U	N	D	L	E	A	S	S	, M	A	I	N	*[4]	E	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-											
17	61-30198-446		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17	61-30198-447		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17	61-30198-448		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17	61-30198-449		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17	61-30198-450		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17	61-30198-453		.	WIRE BUNDLE ASSY, MAIN	*	[4]				N	1
17A	209592		.	GASKET, V33525	*	[3]					4
17B	101A052-3		.	CONNECTOR CAP, V06090 (USED TO CAP CONNECTOR WHEN VIBRATION DETECTOR NOT INSTALLED)						H	2
18	69-52264-1		.	RACEWAY INSTL						C	1
19	69-52264-2		.	RACEWAY							1
20	BACS12CB3-9		.	SCREW							2
21	65-64491-1		.	RACEWAY INSTL						C	1
22	65-64491-2		.	RACEWAY							1
23	69-52265-1		.	BRACKET ASSY							1
24	BACS12CB3-9		.	SCREW							2
25	AN960-10L		.	WASHER							2
25A	BACC10BH2RW		.	CLAMP	*	[6]					AR
26	BACC10BH3RW		.	CLAMP	*	[6]					AR
27	BACC10BH4RW		.	CLAMP	*	[6]					AR
28	BACC10BH5RW		.	CLAMP	*	[6]					AR
29	BACC10BH6RW		.	CLAMP	*	[6]					AR
30	BACC10BH7RW		.	CLAMP	*	[6]					AR
31	BACC10BH8RW		.	CLAMP	*	[6]					AR
32	BACC10BH9RW		.	CLAMP	*	[6]					AR
33	BACC10BH12RW		.	CLAMP	*	[6]					AR
34	BACC10BH14RW		.	CLAMP	*	[6]					AR
35	BACC10BH15RW		.	CLAMP	*	[6]					AR
36	BACS12CB3		.	SCREW							AR
37	NAS679A3W		.	NUT							AR
37A	BACC10HS(-)		.	CLAMP (REPLS BACC10EP(-))	*	[6]					7
37A	BACC10EP(-)		.	CLAMP (REPLD BY BACC10HS(-))	*	[6]					7
37B	BACC10GE4		.	CLAMP	*	[5]					1
37D	AN960D10L		.	WASHER	*	[5]					3
37F	NAS43HT3-80		.	SPACER	*	[5]					1
37H	NAS1801-3-30		.	SCREW	*	[5]					1
37J	NAS1801-3-8		.	SCREW	*	[5]					3
37K	NAS1801-04-5		.	SCREW	*	[5]					4
37P	BACN10NW1		.	NUT	*	[5]					4
38	BACB30LM6U-9		.	BOLT							1
38A	AN6C-11		.	BOLT							AR
38B	NAS566-39		.	BOLT							1
39	AN960C616		.	WASHER							2
39A	AN960-616L		.	WASHER							AR

737
POWERPLANT BUILDUP



FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1117-40	BACN10JC6		.								1
40A	NAS679AGW		.								AR
41	BACS12CB3-8		.								3
42	AN960D10L		.								5
43	NAS43HT3-56		.								3
44	BACS12CB3-32		.								2
45	BACS12CB3-24		.								1
46	NAS43HT3-48		.								1
47	BACS12CB3-22		.								1
48	BACT12C20		.								2
49	NAS43HT3-24		.								1
50	NAS43HT3-32		.								2
51	BACC10HS05		.								1
51	BACC10EP5		.								1
52	BACC10HS04		.								1
52	BACC10EP4		.								1
53	NAS43HT3-16		.								1
54	BACC10DK2		.								1
55	BACS12CB3-7		.								2
58	MS27294-3		.							E	1
59	BACS12CB04		.								AR
60	BACN10JC3		.								AR
61	AN960PD10L		.								AR
-62	61-30194-003		.							B-D G-OQ	1

*[1] FOR SPARES, REFER TO 65-52971-1. 65-52971-5001 THRU -5025 ARE VARIABLE INSTALLATIONS OF 65-52971-1.

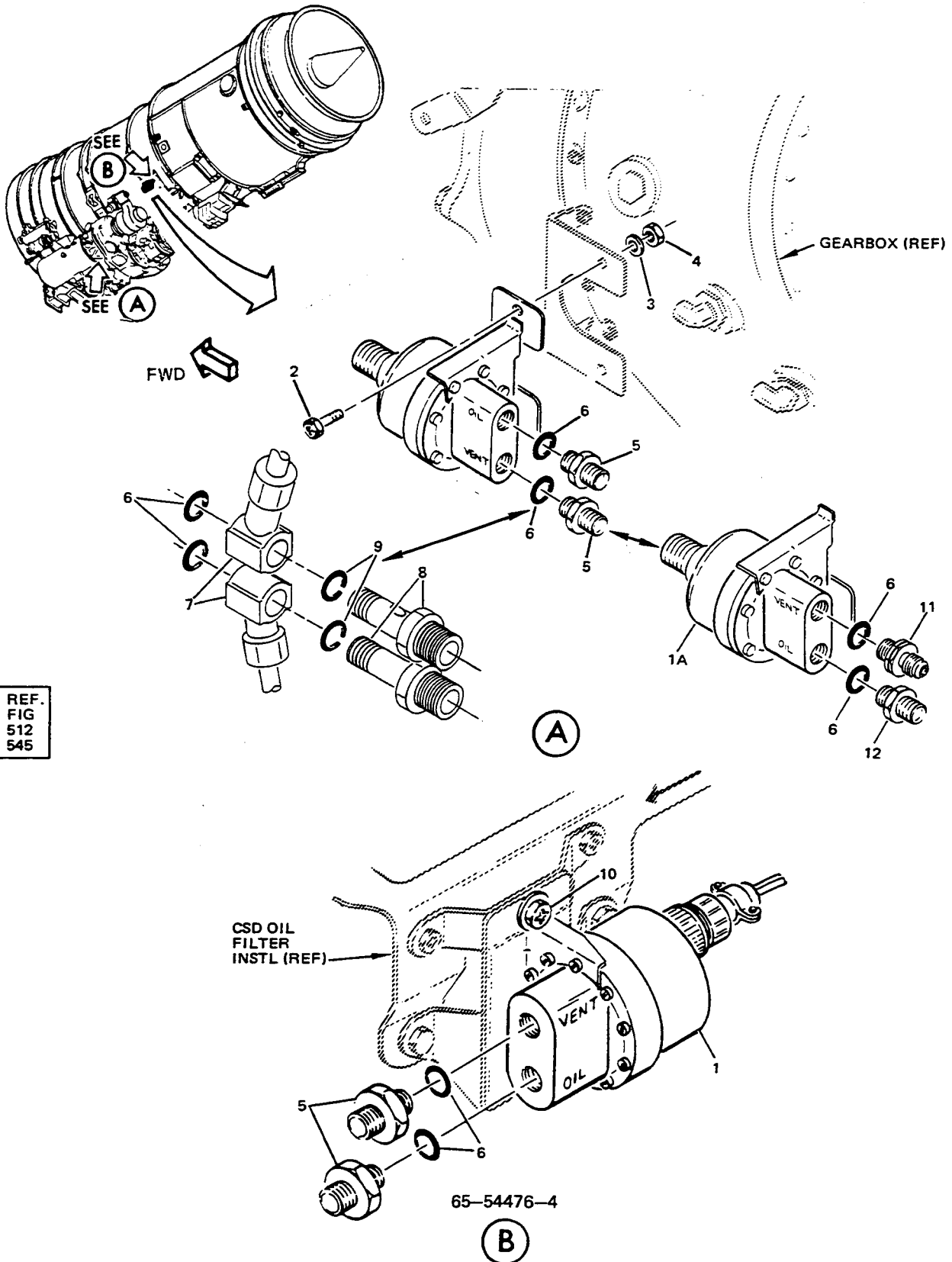
*[2] SUPPLIED WITH VORTEX DISSIPATOR INSTALLATION KIT

*[3] SUPPLIED WITH KIDDE FIRE DETECTOR 10-61096-19 AND -115

*[4] LIMITED USAGE

*[5] POST SB 80-1004 THRU 80-1008, 80-1011

*[6] CLAMP SIZE DETERMINED ON INSTALLATION.



REF.
FIG
512
545

Engine Differential Oil Pressure Switch Installation
Figure 1118

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1118-	65-54476-1		SWITCH INSTL, ENG DIFFERENTIAL OIL PRESSURE *[1]							A	RF
	65-54476-2		SWITCH INSTL, ENG DIFFERENTIAL OIL PRESSURE							B	RF
	65-54476-3		SWITCH INSTL, ENG DIFFERENTIAL OIL PRESSURE							C	RF
	65-54476-4		SWITCH INSTL, ENG DIFFERENTIAL OIL PRESSURE *[1]							D	RF
1	42D107A1		. SWITCH, (V09049)(BOEING 10-3269-8) (POST SB 79-1004)							ABD	1
1A	69-72688-1		. SWITCH ASSY (POST SB 79-1004)							C	1
2	NAS1801-3-8		. SCREW (REPLS BACSL2CB3-8)							ABC	2
2	BACSL2CB3-8		. SCREW (REPLD BY NAS1801-3-8)							ABC	2
3	AN960-10		. WASHER							ABC	2
4	NAS679A3W		. NUT							ABC	2
5	MS21916-6-4		. REDUCER UNION (POST SB 79-1004)							AD	2
6	BACP11PB4		. O-RING								2
7	66-19214-1		. ELBOW - 75 DEGREE							B	2
8	69-45458-1		. BOLT							B	2
9	BACP11P11		. O-RING							B	2
10	NAS1801-3-7		. SCREW							D	2
10	NAS1801-3-8		. SCREW (POST SB 79-1004)								2
-10A	AN960D10L		. WASHER (POST SB 79-1004)								2
11	MS21900-4		. ADAPTER							C	1
12	MS21902-4		. UNION							C	1

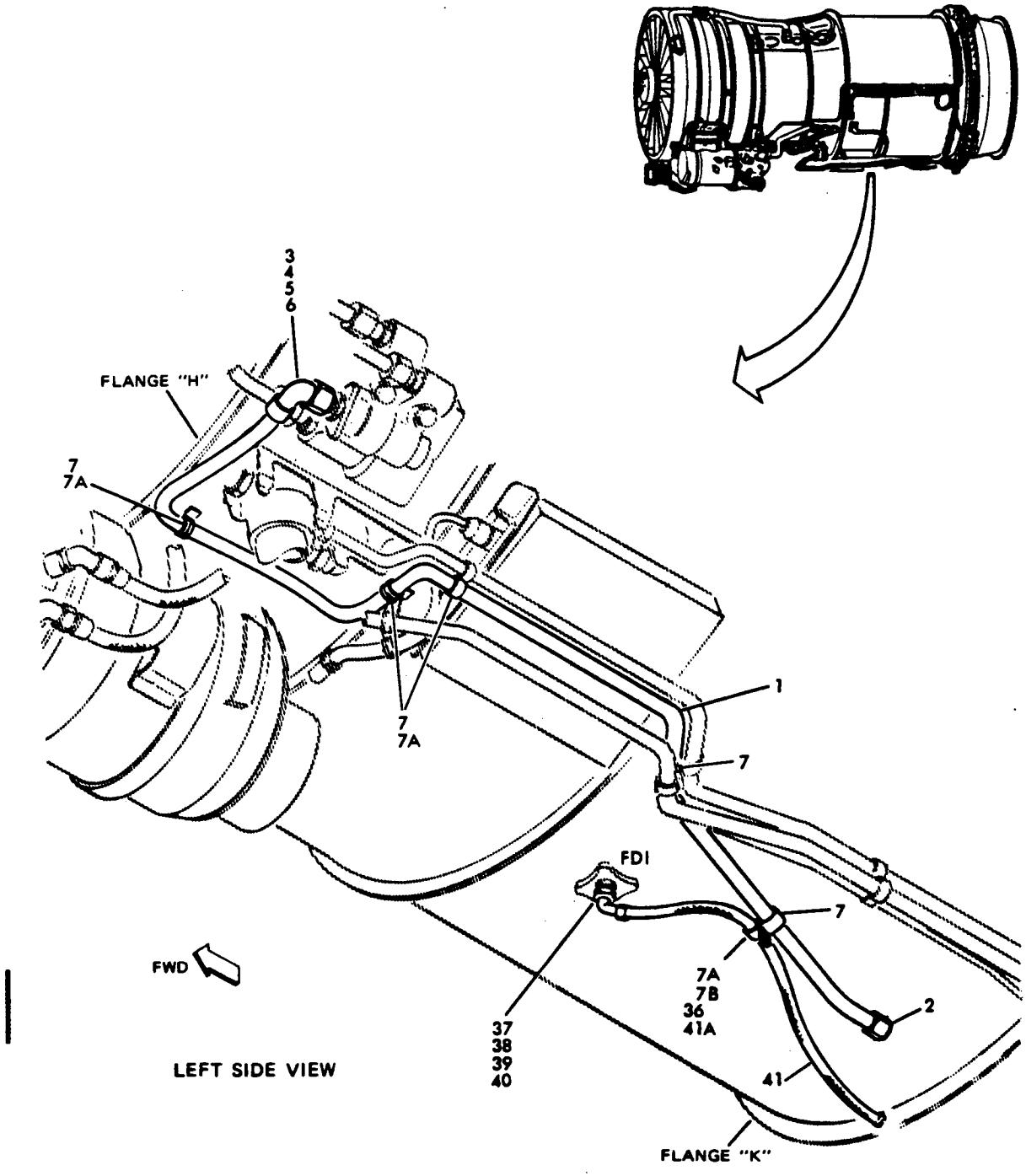
-ITEM NOT ILLUSTRATED

*[1] DIFFERENTIAL OIL PRESSURE SWITCH RELOCATED FROM CSD OIL FILTER TO GEARBOX PER MC 3550-6, -7

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

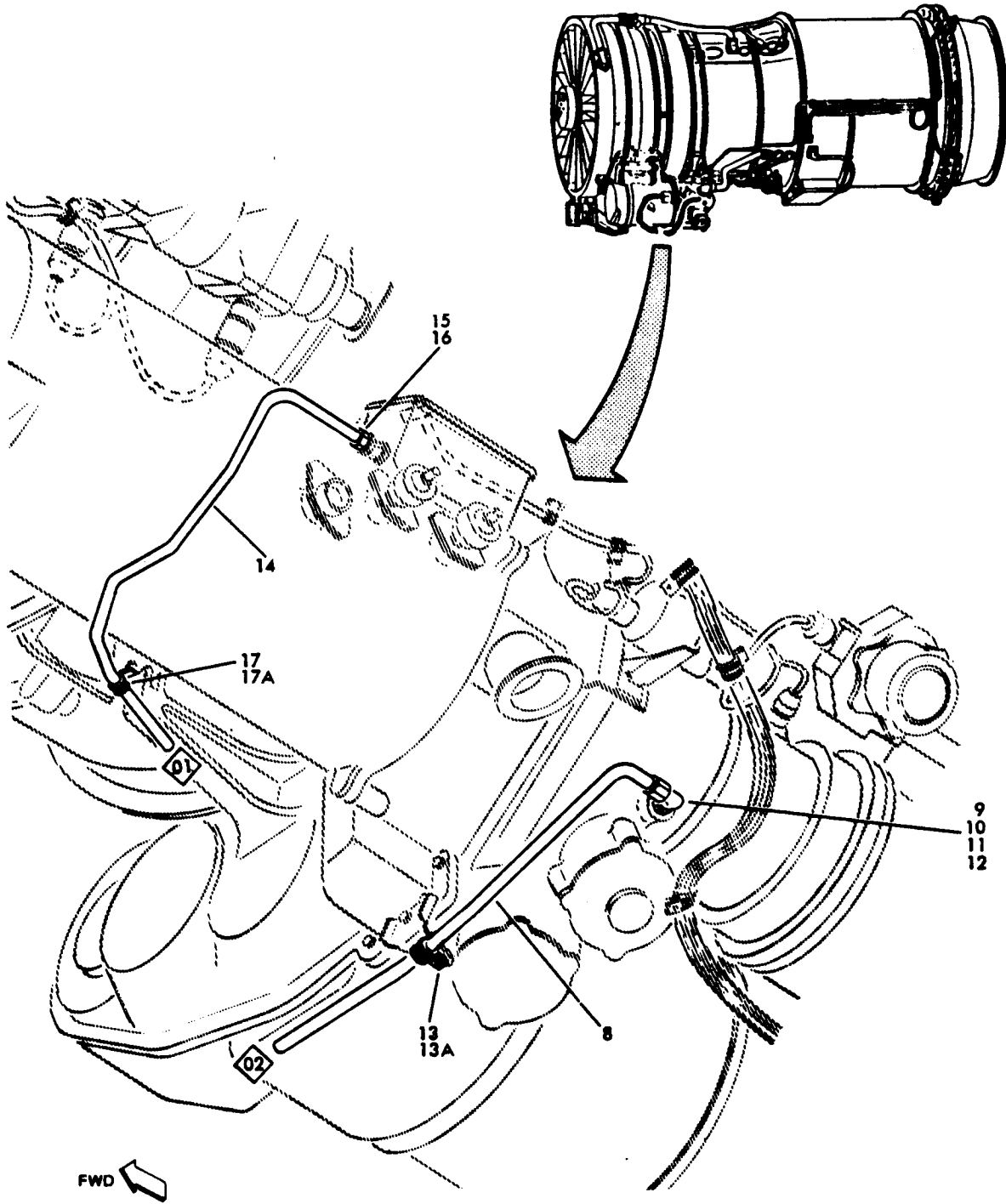


REF.
FIG
553

Sep 10/72

Engine Drains
Figure 1119 (Sheet 1)

71-03-01
Page 1128E



Engine Drains
Figure 1119 (Sheet 2)

Feb 15/69

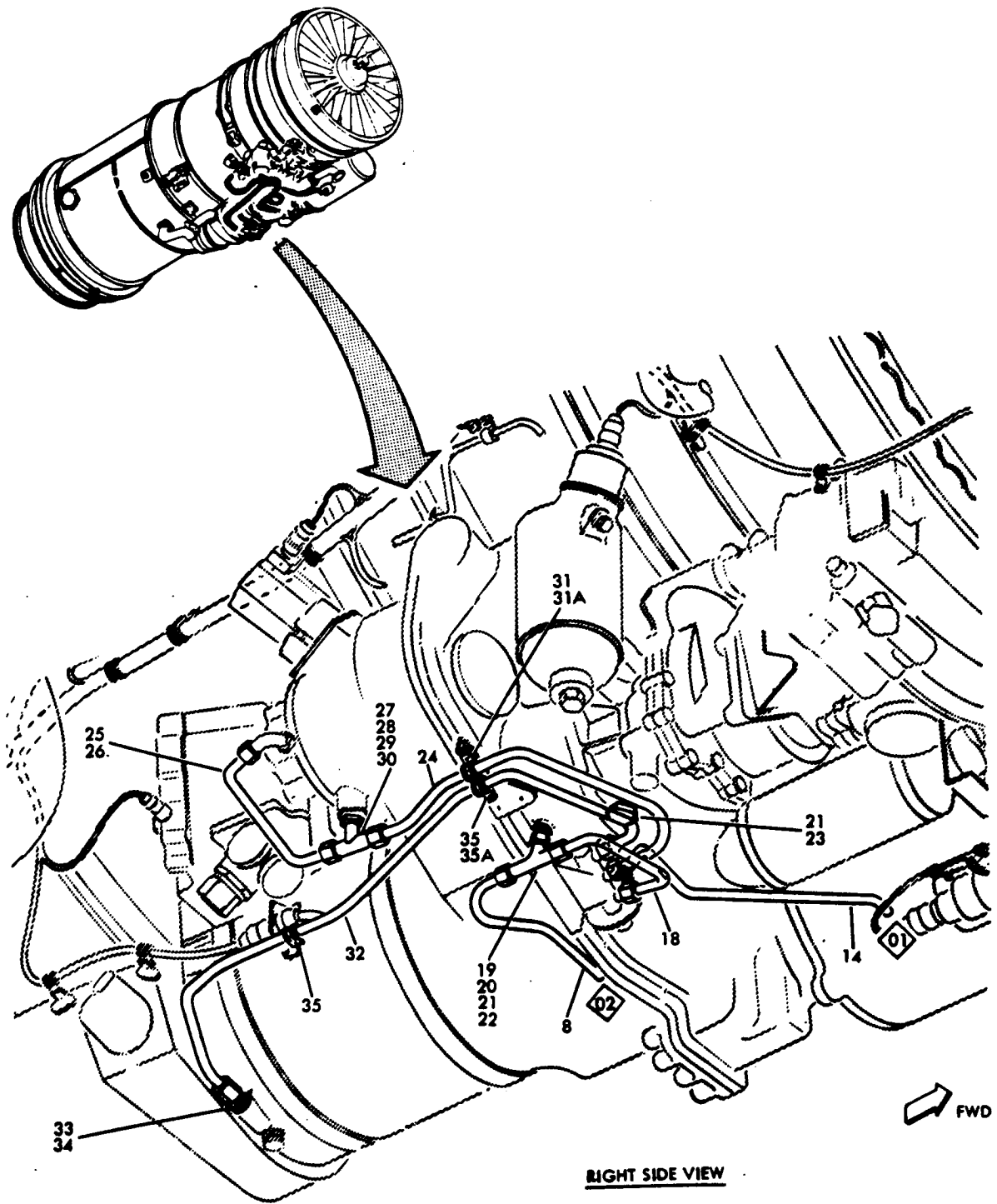


FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1119-	65-55475-1		PLUMBING INSTL, ENGINE DRAIN SYSTEM							A	RF
	65-55475-21		PLUMBING INSTL, ENGINE DRAIN SYSTEM							B	RF
	65-55475-26		PLUMBING INSTL, ENGINE DRAIN SYSTEM							C	RF
	65-55475-26		PLUMBING INSTL, ENGINE DRAIN SYSTEM (POST SB 71-1179, 71-1198)							C	RF
	65-55475-28		PLUMBING INSTL, ENGINE DRAIN SYSTEM							D	RF
	65-55475-32		PLUMBING INSTL, ENGINE DRAIN SYSTEM							E	RF
	65-55475-33		PLUMBING INSTL, ENGINE DRAIN SYSTEM							F	RF
	65-55475-34		PLUMBING INSTL, ENGINE DRAIN SYSTEM							G	RF
			P&D VALVE DRAIN								
1	65-55475-27		. TUBE ASSY *[2]								
1	65-55475-27		. TUBE ASSY *[3]*[5]							C	1
1	65-55475-11		. TUBE ASSY *[1] (LIMITED USAGE)							ABD-G	1
2	MS21902-12		. UNION								1
3	MS21908-12		. ELBOW *[3] (LIMITED USAGE)								1
4	MS28777-12		. RING, BACKUP *[3] (LIMITED USAGE)								1
5	AN6289-12		. NUT *[3] (LIMITED USAGE)								1
6	BACP11PB12		. PACKING, O-RING *[3](LIMITED USAGE)								1
7	BACC10HS12		. CLAMP *[6]								5
7	BACC10EP12		. CLAMP *[6] (LIMITED USAGE)								5
7	BACC10DDC12		. CLAMP *[6]								5
7A	BACS12CB3-8		. SCREW								4
7B	NAS679A3W		. NUT								1
			STARTER PAD DRAIN								
8	65-55475-25		. TUBE ASSY								1
9	MS21926-6		. ELBOW								1
10	MS28777-6		. RING, BACKUP								1
11	AN6289-6		. NUT								1
12	BACP11PB6		. PACKING, O-RING								1
13	BACC10HS06		. CLAMP *[6]								1
13	BACC10EP6		. CLAMP *[6]								1
13	BACC10DDC6		. CLAMP *[6]								1
13A	BACS12CB3-7		. SCREW								1
			ENGINE OIL TANK DRAIN								
14	65-55475-4		. TUBE ASSY								1
15	MS21902-6		. UNION								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1119-16	BACP11PB6		.								1
17	BACC1OHS06		.								1
17	BACC1OEP6		.								1
17	BACC1ODDC6		.								1
17A	BACS12CB3-7		.								1
			FUEL PUMP PAD DRAIN								
18	65-55475-5		.								1
19	2-02302		.								1
19	118-10555		.								1
			2-02302)								
20	AN6289-6		.								1
21	BACP11PB6		.								2
22	MS28777-6		.								1
23	AN814-6CL		.								1
			HYDRAULIC PUMP DRAIN								
24	65-55475-24		.								1
25	65-55475-8		.						ACF		1
25	65-55475-31		.						EG		1
25	65-55475-29		.						D		1
26	65-55475-22		.						B		1
27	MS21909-6		.								1
28	MS28777-6		.								1
29	AN6289-6		.								1
30	NAS1612-6		.								1
31	BACC1OHS06		.								1
31	BACC1OEP6		.								1
31	BACC1ODDC6		.								1
31A	BACS12CB3-7		.								1
			CONSTANT SPEED DRIVE DRAIN								
32	65-55475-23		.						A-DG		1
32	65-55475-30		.						EF		1
33	MS21902-6		.								1
34	BACP11PB6		.								1
35	BACC1OHS06		.								2
35	BACC1OEP6		.								2
35	BACC1ODDC6		.								2
35A	BACS12CB3-7		.								2

737

POWERPLANT BUILDUP

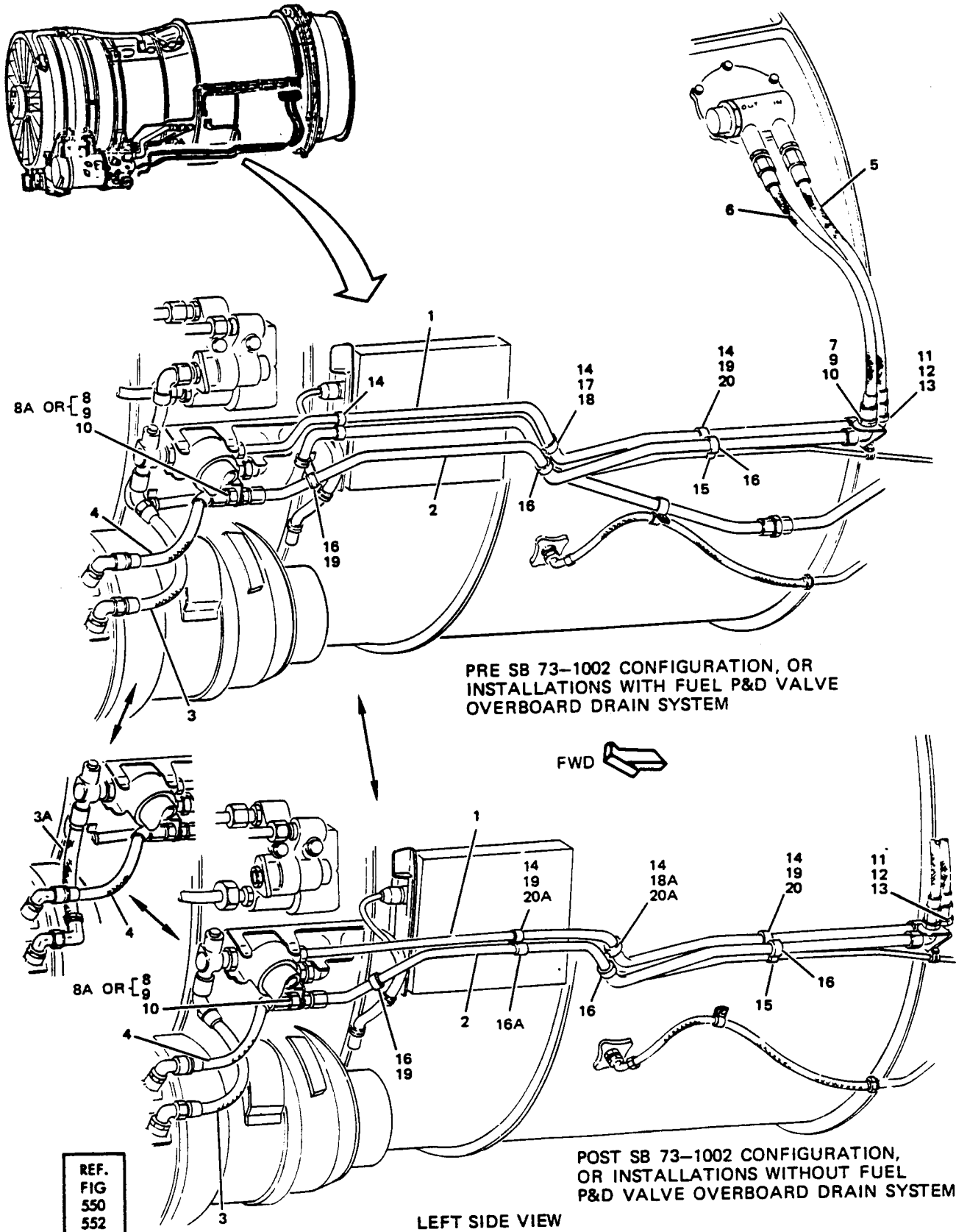


OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1119-			COMBUSTION CHAMBER DRAIN								
36	BACC10HS14		. CLAMP *[6]								1
36	BACC10EP14		. CLAMP *[6]								1
36	BACC10DDC14		. CLAMP *[6]								1
37	MS21926-6		. ELBOW								1
38	MS28777-6		. RING, BACKUP								1
39	AN6289-6		. NUT								1
40	BACP11PB6		. PACKING, O-RING								1
41	BACH30AS06F 0201		. HOSE ASSY (LIMITED USAGE)								1
41	BACH5P0184CCF		. HOSE ASSY (SB 73-1002) (LIMITED USAGE)								1
41A	NAS43HT3-56		. SPACER (SB 73-1002)								1
			STRUT DRAIN *[4]								
42	65-55475-13		. TUBE ASSY (LIMITED USAGE)								1
43	65-55475-14		. TUBE ASSY (LIMITED USAGE)								1
44	65-55475-16		. TUBE ASSY (LIMITED USAGE)								1
45	BACC10HS06		. CLAMP *[6]								12
45	BACC10EP6		. CLAMP *[6]								12
45A	NAS43HT3-40		. SPACER								2
45B	BACS12CB3-7		. SCREW								11
45C	BACS12CB3-20		. SCREW								2
45D	NAS679A3W		. NUT								AR
46	65-55475-802		. TUBE ASSY (LIMITED USAGE)								1
47	65-55475-803		. TUBE ASSY (LIMITED USAGE)								1
48	AN924-6		. NUT (LIMITED USAGE)								4
49	MS21902-6		. UNION (LIMITED USAGE)								2
50	MS21908-6		. ELBOW (LIMITED USAGE)								1
51	AN960-916		. WASHER								4
52	MS21909-6		. TEE (LIMITED USAGE)								1
53	MS21924-6		. UNION (LIMITED USAGE)								2
54	BACH30AV06F 0163		. HOSE (LIMITED USAGE)								1
55	DELETED										
56	DELETED										
57	69-48904-1		. DRAIN FITTING (LIMITED USAGE)								1
58	AN924-6		. NUT (LIMITED USAGE)								1
59	AN960-916		. WASHER								1
60	BACS12CB3-6		. SCREW								1
61	NAS679A3W		. NUT								1
62	AN96010L		. WASHER								1

OVERHAUL MANUAL

- *[1] PRE SB'S 71-1032, 71-1064, 71-1071, 71-1092, 71-1093, 71-1100, 71-1101, 71-1115, 71-1118, 71-1125, 71-1140, 71-1151, 71-1179, 71-1198, 71-1278
- *[2] POST SB'S 71-1032, 71-1064, 71-1071, 71-1092, 71-1093, 71-1100, 71-1101, 71-1115, 71-1118, 71-1125, 71-1278
- *[3] NOT USED WHEN SB 73-1002 INCORPORATED
- *[4] USED ON POWER PLANTS EQUIPPED WITH PNEUMATICALLY-OPERATED THRUST REVERSERS
- *[5] POST SB 71-1140, 71-1151
- *[6] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC()
CLAMPS



PRE SB 73-1002 CONFIGURATION, OR
 INSTALLATIONS WITH FUEL P&D VALVE
 OVERBOARD DRAIN SYSTEM

FWD 

POST SB 73-1002 CONFIGURATION,
 OR INSTALLATIONS WITHOUT FUEL
 P&D VALVE OVERBOARD DRAIN SYSTEM

LEFT SIDE VIEW

REF.
 FIG
 550
 552

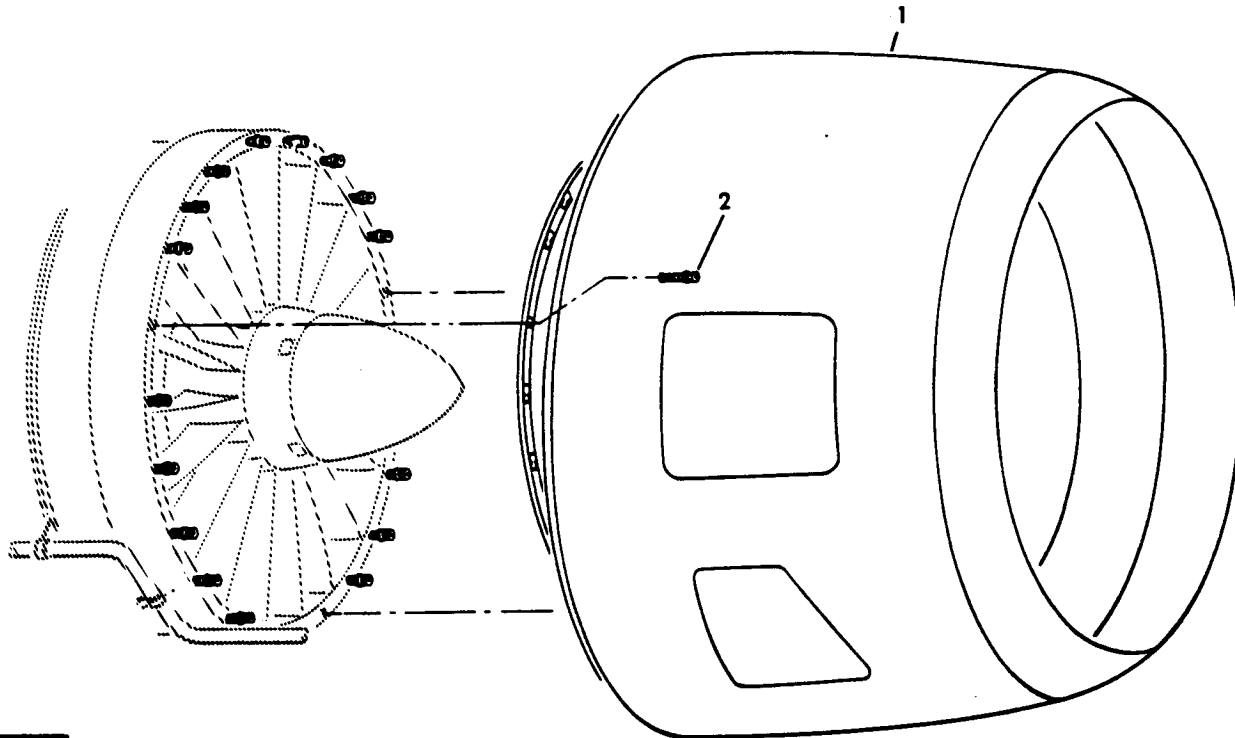
737
POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1120-	65-55749-1									A	RF
	65-55749-8									B	RF
	65-55749-9									C	RF
1	65-55749-4										2
1	65-55749-6										2
2	65-55749-5										2
3	BACH30AU08F 0116									A	1
3	BACH5R0116DUF									B	1
3A	BACH30AU08F 0116									C	1
4	BACH30AV10F 0114									AC	1
4	BACH5S0074EVF									B	1
5	BACH5R0173DDF										1
5	BACH30AS08F 0173										1
6	BACH5R0173EEF										1
6	BACH30AS10F 0173										1
7	MS21908-10										1
8	MS21924-10									AC	1
8A	MS21907-10									B	1
9	AN960-1416									B	2
10	AN924-10										2
11	MS21908-8										1
12	AN924-8										1
13	AN960-1216										2
14	BACC10HS08										3
14	BACC10EP8										3
14	BACC10DDC8										3
15	BACC10HS12										1
15	BACC10EP12										1
15	BACC10DDC12										1
16	BACC10HS10										4
16	BACC10EP10										4
16	BACC10DDC10										4
16A	BACC10HS10										2
16A	BACC10EP10										2
17	NAS43HT3-64										2

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY												
			1	2	3	4	5	6	7														
1120-18	BACSI2CB3-26		.	S	C	R	E	W			2												
18A	BACSI2CB3-9		.	S	C	R	E	W	(P	O	S	T	S	B	7	3	-	1	0	0	2)
19	BACSI2CB3-8		.	S	C	R	E	W			2												
20	NAS679A3W		.	N	U	T					A	R											
20A	BACN10JC3		.	N	U	T	(P	O	S	T	S	B	7	3	-	1	0	0	2)		

- *[1] SAME AS 65-55749-1 EXCEPT HOSE ASSY BACH30AU08F0116 IS RE-POSITIONED (TURNED END FOR END)
- *[2] REPOSITION HOSE ASSY BACH5R0116DUF (TURN END FOR END) FOR LIMITED EFFECTIVITIES
- *[3] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC() CLAMPS



REF.
 FIG
 562
 567

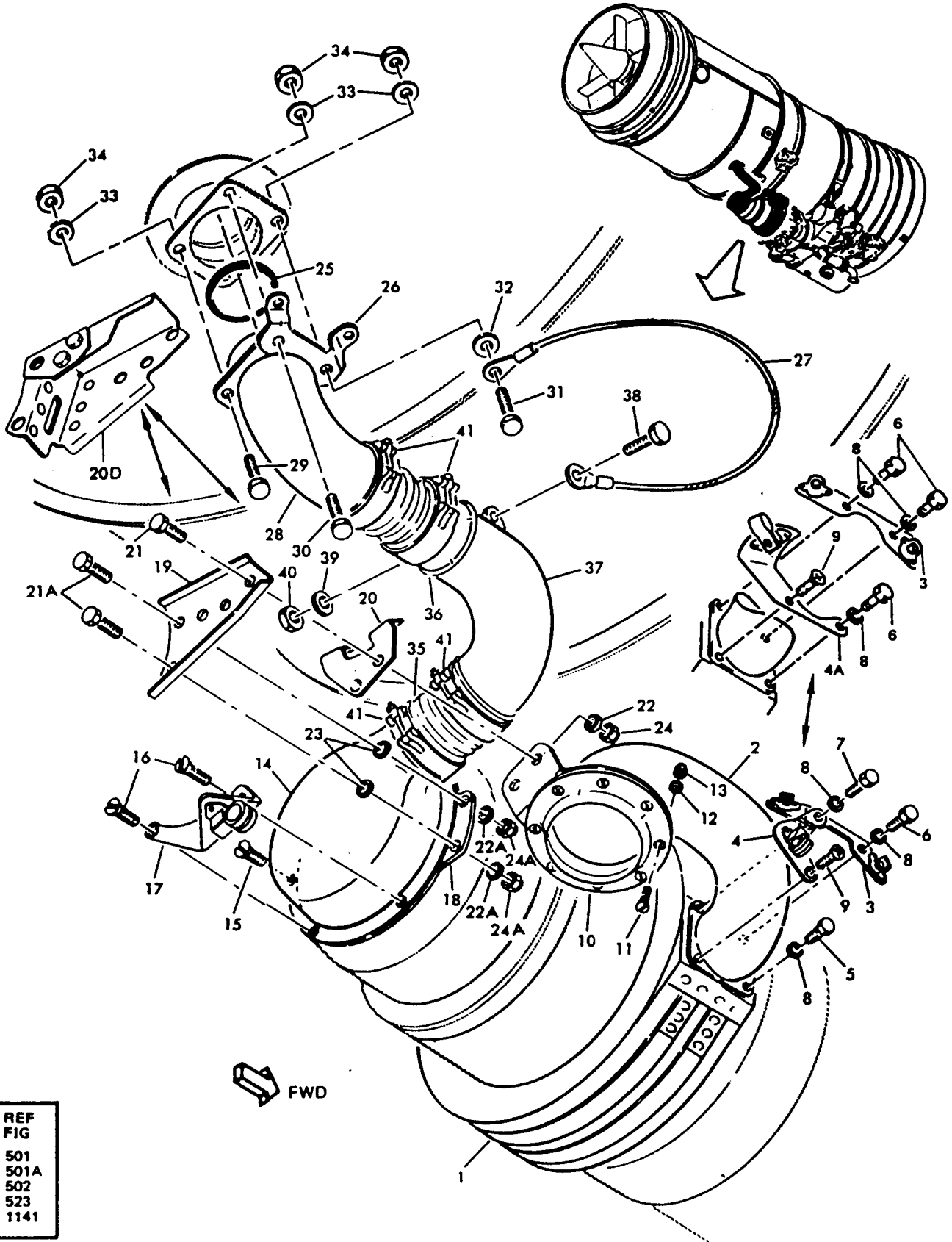
Nose Cowl Installation
 Figure 1121

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1121	65-55750-1										
	65-55750-2										
1	65-60529-767										1
1	65-60529-787										1
1	65-60529-1499										1
1	65-60529-673										1
1	65-60529-771										1
1	65-68621-1										1
1	65-60529-768										1
1	65-60529-788										1
1	65-60529-1500										1
1	65-60529-674										1
1	65-60529-772										1
1	65-68621-27										1
2	MS9148-08										23

OVERHAUL MANUAL

- *[1] USED ON ENGINES WITH BOOM TYPE VORTEX DISSIPATOR.
- *[2] PRE SB'S 71-1064, 71-1067, 71-1093. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[3] PRE SB'S 71-1063, 71-1065, 71-1071, 71-1092, 71-1105, 71-1115, 71-1140, 71-1142, 71-1147. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[4] PRE SB'S 71-1083, 71-1100. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[5] PRE SB 71-1101, 71-1032. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[6] PRE SB 71-1118, 71-1125, 71-1125R1, 71-1125R2. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[7] PRE SB 71-1121, 71-1128, 71-1198. REFER TO FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION.
- *[8] PRE SB 71-1151 SEE FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION
- *[9] PRE SB 71-1179 SEE FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION
- *[10] PRE SB 71-1278 SEE FIG. 1143 FOR POST SERVICE BULLETIN CONFIGURATION

NOTE: SEE FIG. 1143 FOR FAT LIP NOSE COWL INSTALLATION



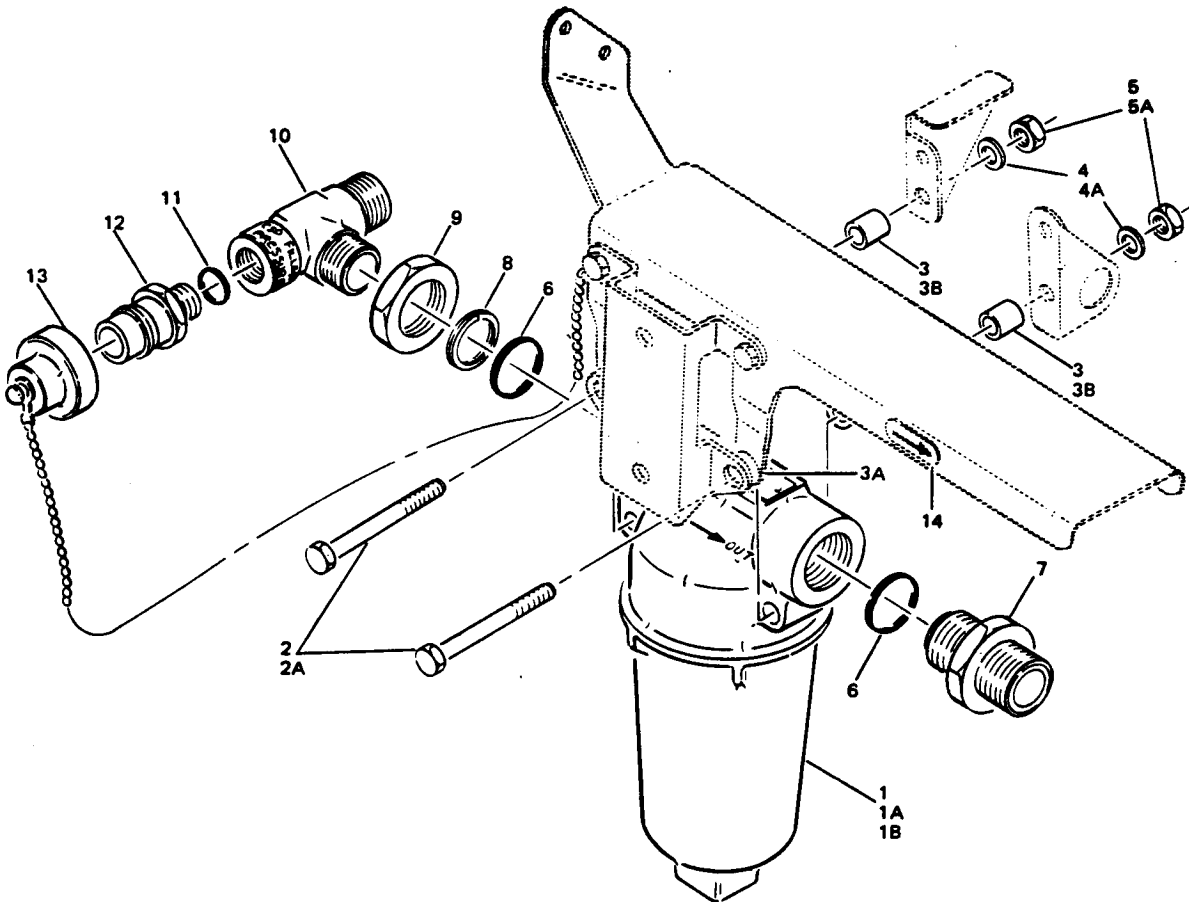
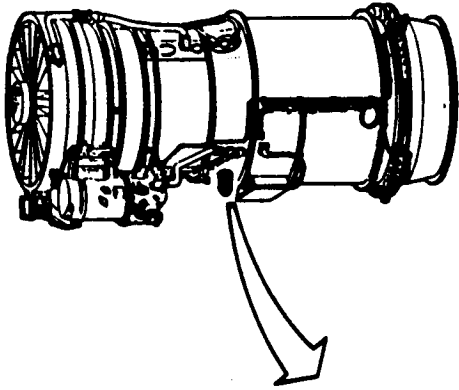
REF
FIG
501
501A
502
523
1141

AC Generator Cooling Duct Installation
 Figure 1122

Sep 5/87

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1122-	65-56646-1		DUCT INSTL, AC GENERATOR COOLING							A	RF
	65-56646-2		DUCT INSTALLATION, AC GENERATOR COOLING							B	RF
	65-56646-3		DUCT INSTALLATION, AC GENERATOR COOLING							C	RF
1	65-22429-15		. COLLECTOR RING (LIMITED USAGE)							A	1
1	65-22429-19		. COLLECTOR RING (LIMITED USAGE)							A	1
1	65-22429-22		. COLLECTOR RING							BC	1
2	65-46505-1		. FITTING							A	1
2	65-46505-3		. FITTING, EXHAUST DUCT							BC	1
3	69-45402-1		. BRACKET ASSY							A	1
4	69-52217-3		. BRACKET ASSY							A	1
4A	69-71727-8		. BRACKET ASSY							C	1
5	NAS1103-5		. BOLT							AB	1
6	NAS1103-6		. BOLT							AB	1
6	NAS1103-6		. BOLT							C	3
7	NAS1103-7		. BOLT							AB	1
8	AN960C10L		. WASHER								3
9	BACB30LU3-6		. SCREW								1
10	69-31696-1		. SEAL							A	1
11	BACB30FL3-2		. BOLT							A	8
11	BACB30LU3-2		. BOLT							BC	8
12	AN960-10L		. WASHER								8
13	NAS679A3W		. NUT							A	8
13	BACN10JC3		. NUT							BC	8
14	938D844-3		. CONNECTOR ASSY, AIR BLAST, V83843							A	1
14	332A1025-2		. ADAPTER, GENERATOR TO AIR DUCT (OPT TO 938D844-3)							A	1
14	65C20874-1		. CONNECTOR							BC	1
15	AN501AD10-7		. SCREW							A	3
15	NAS1802-3D7		. SCREW							BC	3
16	AN501AD10-8		. SCREW							A	2
16	NAS1802-3D8		. SCREW							BC	2
17	69-52217-2		. BRACKET ASSY							A	1
18	69-43285-1		. PLATE, DUCT SUPPORT							A	1
19	69-43284-6		. BRACKET							A	1
20	69-43284-5		. BRACKET ASSY							A	1
20D	69-43284-7		. BRACKET ASSY							BC	1
20A	69-43284-8		. . BRACKET							BC	1
20B	69-43284-9		. . BRACKET							BC	1
20C	69-43284-10		. . BRACKET							BC	1
21	NAS1103-3		. BOLT								1
21A	NAS1103-3		. BOLT								2

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1122											
22	AN960C10L		.	W	A	S	H	E	R		1
22A	AN960C10L		.	W	A	S	H	E	R		2
23	AN960C10		.	W	A	S	H	E	R	A	2
24	NAS679A3W		.	N	U	T				A	1
24	BACN10JC3		.	N	U	T				BC	1
24A	NAS679A3W		.	N	U	T				A	2
24A	BACN10JC3		.	N	U	T				BC	2
25	400922		.	O	-	R	I	N	G	S	E
26	69-45492-1		.	B	R	A	C	K	E	T	A
27	BACJ40A21-8		.	B	O	N	D	I	N	G	J
27	BACJ40A21-6		.	J	U	M	P	E	R		A
28	65-55860-1		.	D	U	C	T	,	C	O	O
28	65-55860-3		.	D	U	C	T				A
29	NAS1104-4		.	B	O	L	T				BC
30	NAS1104-5		.	B	O	L	T				A
31	NAS1104-6		.	B	O	L	T				BC
32	AN960D416L		.	W	A	S	H	E	R		A
33	AN960C416L		.	W	A	S	H	E	R		BC
34	NAS679A4W		.	N	U	T					A
34	BACN10JC4		.	N	U	T					BC
35	BACD40D10-12-30		.	D	U	C	T				A
36	BACD40D10-15-30		.	D	U	C	T				A
37	69-42961-1		.	D	U	C	T				A
37	69-42961-6		.	D	U	C	T				BC
38	NAS1103-2		.	B	O	L	T				
39	AN960D10		.	W	A	S	H	E	R		
40	NAS679A3W		.	N	U	T					
41	BACC10BN262LR		.	C	L	A	M	P			A



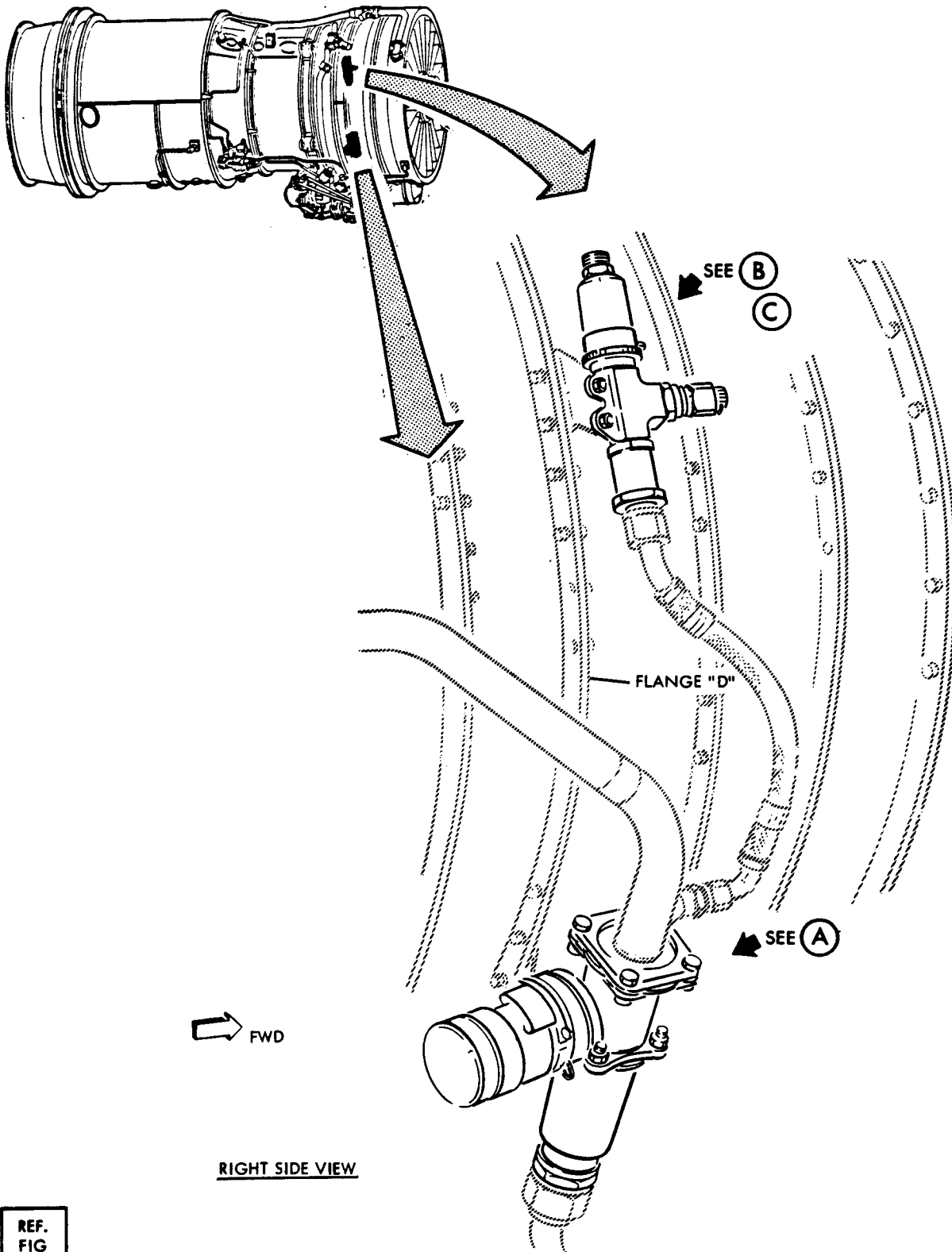
REF.
FIG
503
549

CSD Filter Installation
Figure 1123

737
POWERPLANT BUILDUP

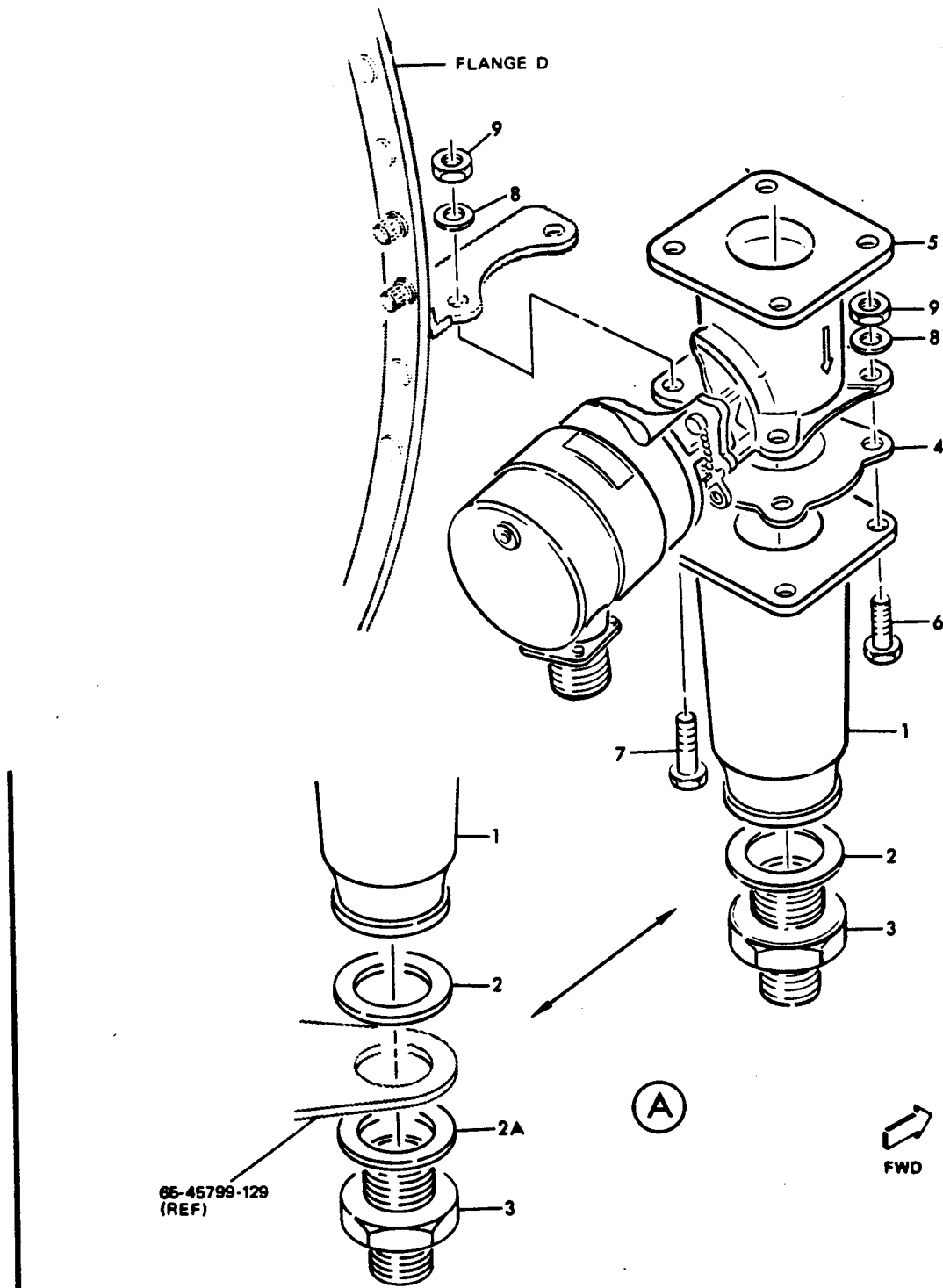
OVERHAUL MANUAL

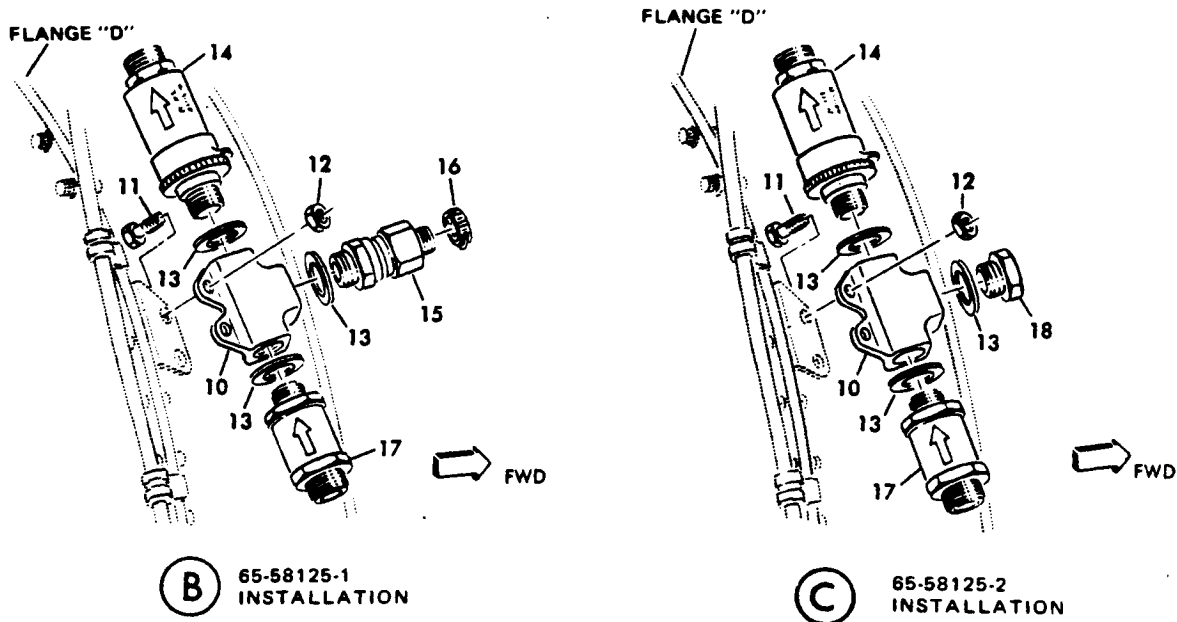
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1123-	65-58122-1									A	RF
	65-58122-2									B	RF
	65-58122-3									C	RF
1	7581418										1
1	7586890										1
1	7500203										1
1A	7513182										1
1A	7583141									C	1
1B	MS29561-228										1
1B	M83248-1-228									C	1
2	NAS1104-36									A	2
2A	BACB30NF4-40									A	2
2A	BACB30NF4-40									BC	2
3	NAS43DD4-28									A	2
3A	NAS43DD4-16									A	2
3A	NAS43DD4-16									BC	2
3B	NAS43DD4-28									BC	2
4	AN960C416L									A	2
4A	AN960C416									A	2
4A	AN960C416									BC	2
5	NAS679A4W									A	2
5A	BACN10JC4									A	2
5A	BACN10JC4									BC	2
6	BACP11PB12										2
7	MS21916-12-8										1
8	MS28777-12										1
9	AN6289-12										1
10	2-02303										1
10	118-10560										1
11	BACP11PB6										1
12	OMP2506-2										1
13	OMP2506-4										1
14	BACM10L001BJZ										1
15	BACM10P6GU										1



REF.
FIG
517
518
534

Nose Cowl Thermal Anti-Icing Valve Installation
Figure 1124 (Sheet 1)





Nose Cowl Thermal Anti-Icing Valve Installation
 Figure 1124 (Sheet 3)

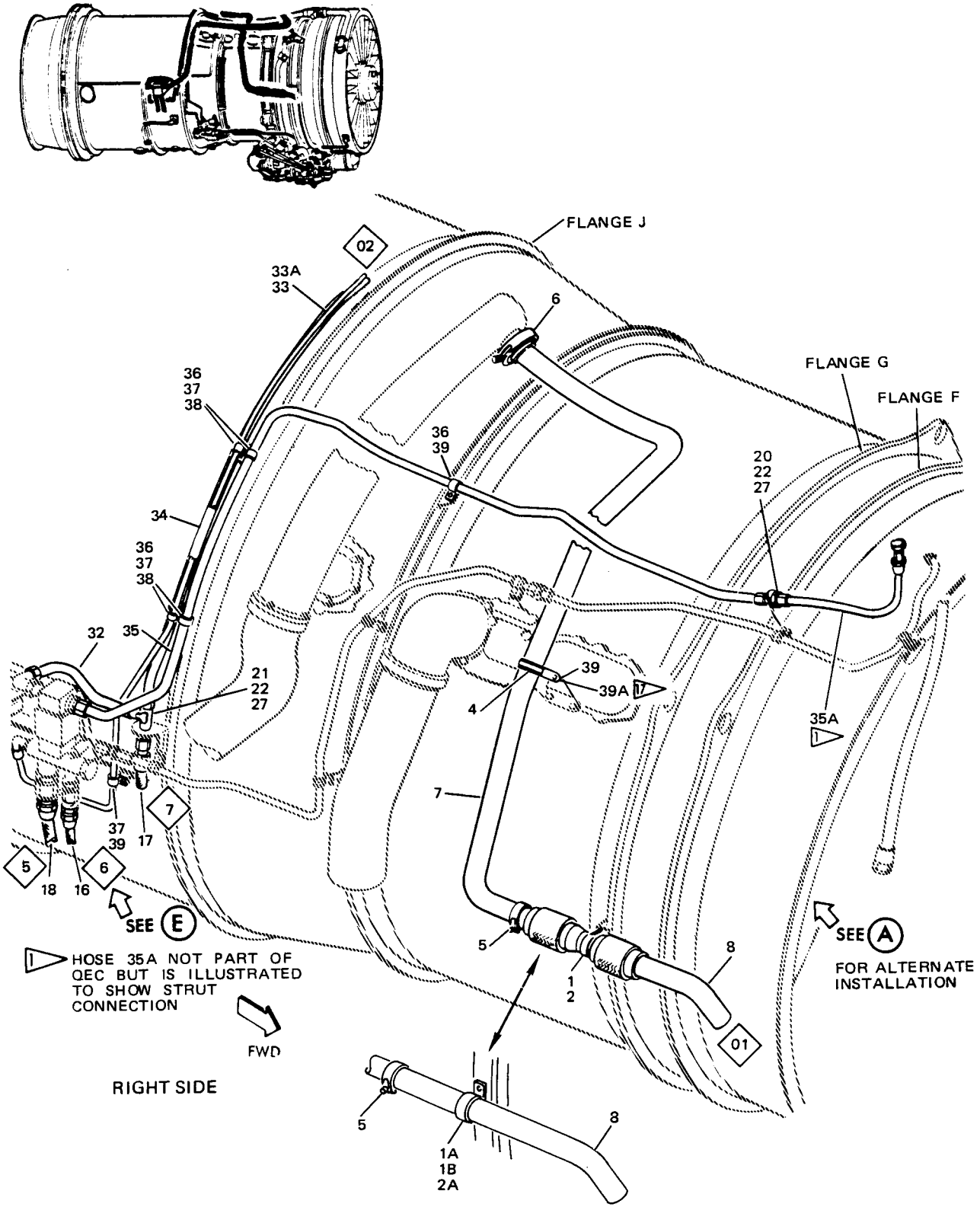
FIG. & INDEX NO.	PART NO.	QEC SUB KIT NO.	AIRLINE PART NO.	1 2 3 4 5 6 7							NOMENCLATURE	USE CODE	UNITS PER ASSY.
1124	65-58125-1										NOSE COWL THERMAL ANTI-ICING VALVE INSTL (REPLD BY 65-58125-3)		
1	65-58125-2										NOSE COWL TAI VALVE INSTL		
1	65-58125-3										NOSE COWL TAI VALVE INSTL		
	392708-1-1	036									. THERMOSTATIC VALVE, V70210		1
	10-60792-1										. THERMOSTATIC VALVE (OPT ON 65-581,-2)		1
2	AN901-24C	036									APPROVED PARTS ARE: 26130105, V80234; 26130105-01, V80234		
2A	AN901-24C	036									. GASKET		1
											. GASKET (USED WITH BRACKET 65-45799-129,-141)		1
3	AN919-28J	036									. REDUCER		1
4	60024	036									. GASKET, V92764		1
4	60023	036									. GASKET, V92764 (OPT ON 65-58125-1,-2)		1

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & INDEX NO.	PART NO.	QEC SUB KIT NO.	AIRLINE PART NO.								NOMENCLATURE	USE CODE	UNITS PER ASSY
				1	2	3	4	5	6	7			
1124-5	229165									. VALVE ASSY, SHUTOFF, V79318 (BOEING 10-60817-10)(USED ON 65-58125-3)(OPT TO 10-60817-8)		1	
5	320115									. VALVE ASSY, SHUTOFF, V79318 (BOEING 10-60817-10)(USED ON 65-58125-3)*[1](OPT TO 10-60817-8)		1	
5	225485									. VALVE ASSY, SHUTOFF, V79318 (BOEING 10-60817-8)(USED ON 65-58125-3)(SB 71-1022)(OPT TO 10-60817-10)		1	
5	129525	036								. VALVE ASSY, SHUTOFF, V79318 (BOEING 10-60817-7)(USED ON 65-58125-1,-2)		1	
6	BACB30LM5U9									. BOLT		2	
6	AN5C10A	036								. BOLT (OPT ON 65-58125-1,-2)		2	
7	AN5C11A	036								. BOLT (USED ON 65-58125-1,-2)		2	
7	BACB30LM5U11	036								. BOLT (USED ON 65-58125-3)		2	
8	AN960C516	036								. WASHER		4	
9	NAS6795C	036								. NUT		4	
10	65-69758-1									DELETED			
10	69-44477-1	036								. TEE		1	
11	BACS12CB3-8	036								. SCREW (USED WITH 69-44477-1)		2	
12	NAS679A3W	036								. NUT		2	
13	AN901-8C	036								. GASKET		3	
14	66-8324-1	036								. FILTER ASSY		1	
15	1112-589971	036								. GROUND SOURCE CONNECTOR, V92003		1	
15	411-00117	036								. GROUND SOURCE CONNECTOR, V92003 (OPT TO 1112-589971)		1	
16	BACC14H8LS	036								. CAP (USED ON 65-58125-1)		1	
17	8C150	036								. CHECK VALVE, V82829		1	
18	BACP20AU-8	036								. PLUG (USED ON 65-58125-2)		1	
19	3-113925-8A									DELETED			
20	BACP11P-B8									DELETED			
21	AN924-8									DELETED			
22	69-55493-5									DELETED			
23	BAC27DPP17									DELETED			
24	BACS12CB3-10									DELETED			

*[1] REWORKED BY VENDOR SERVICE BULLETIN 75-28.

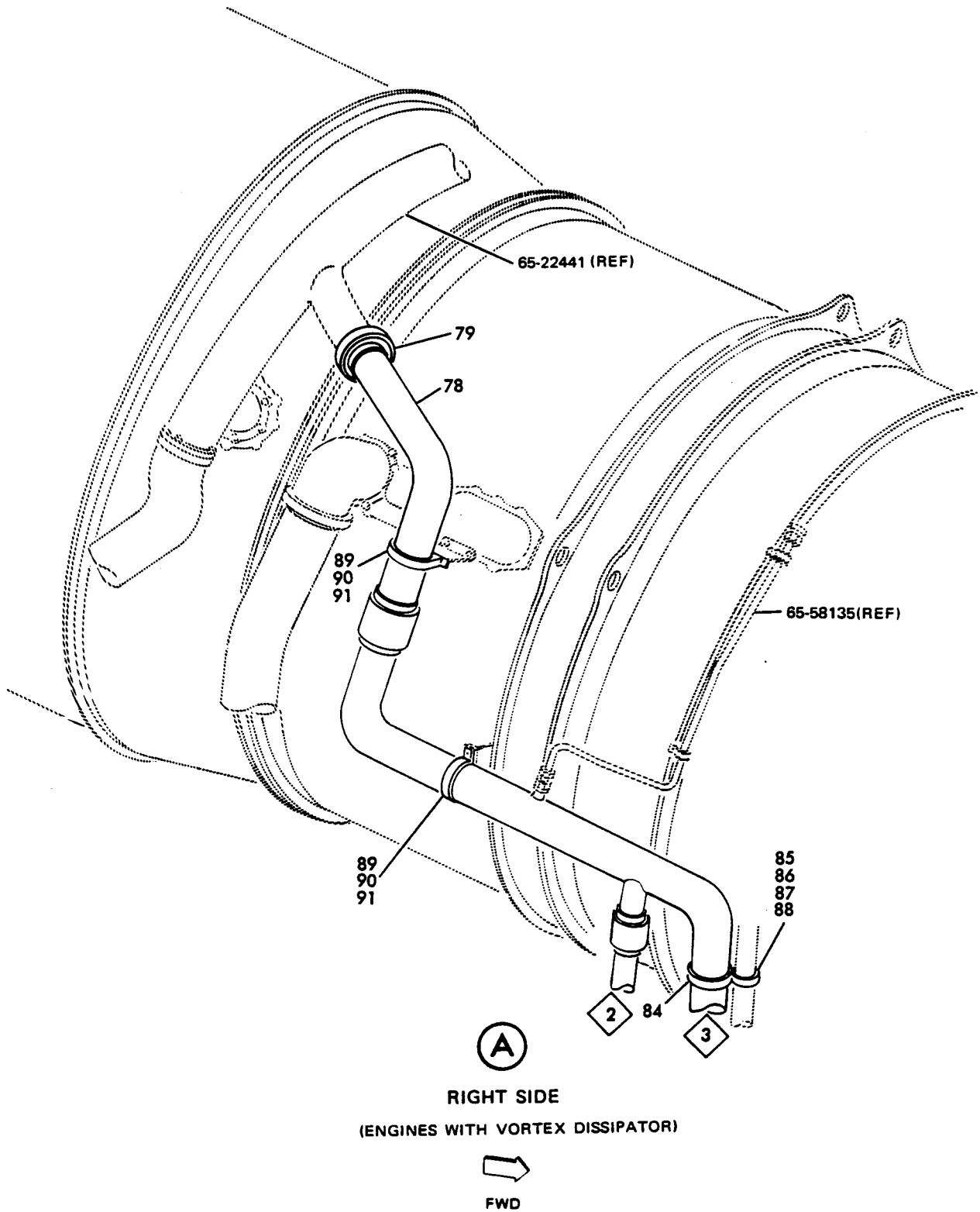


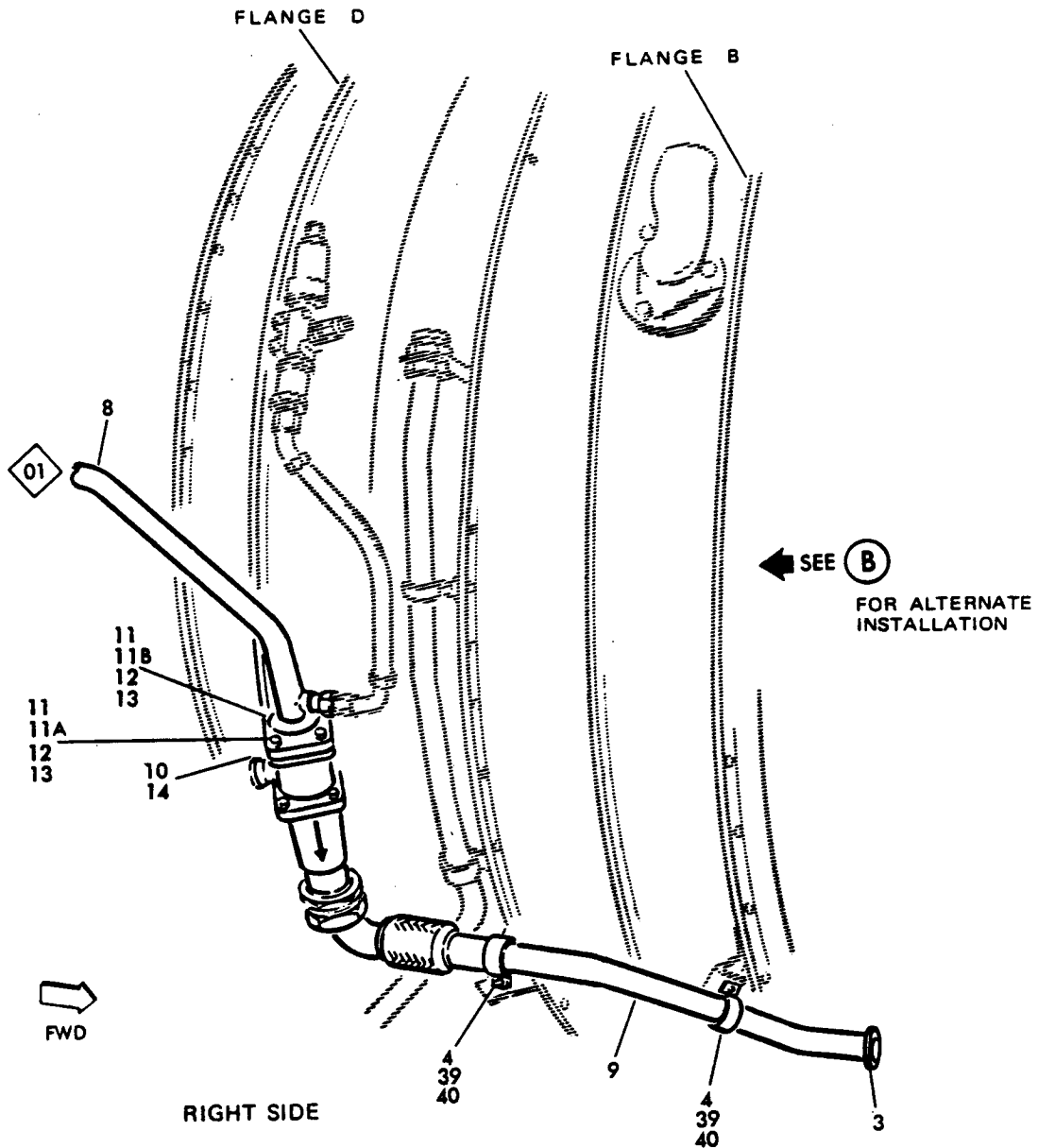
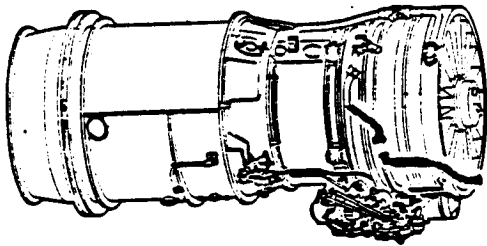
HOSE 35A NOT PART OF QEC BUT IS ILLUSTRATED TO SHOW STRUT CONNECTION

FOR ALTERNATE INSTALLATION

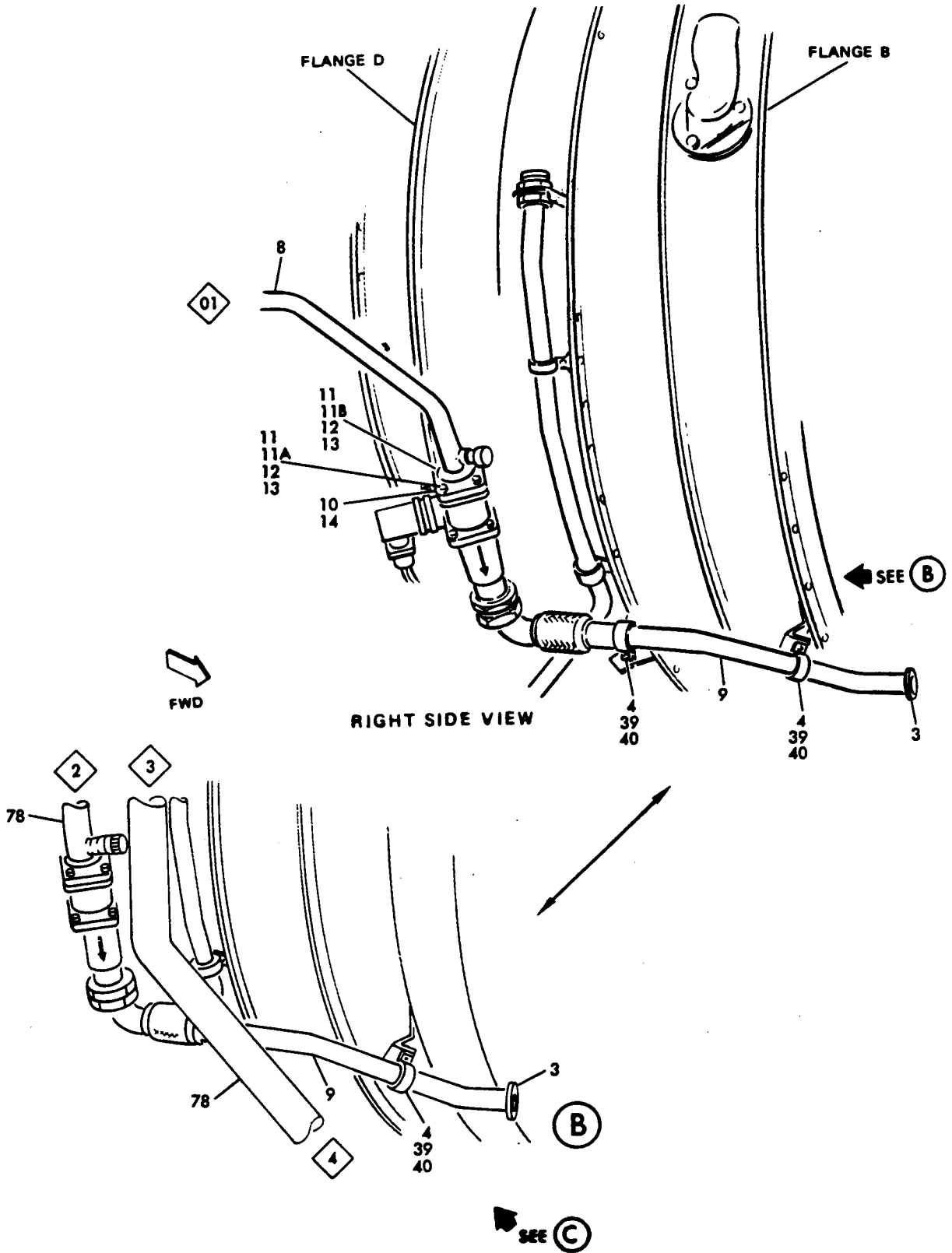
INSTALL SHIM (9A) AT THIS POSITION ONLY IF NECESSARY TO PREVENT PRELOAD CONDITIONS ON TAI DUCT (7).

Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 1)

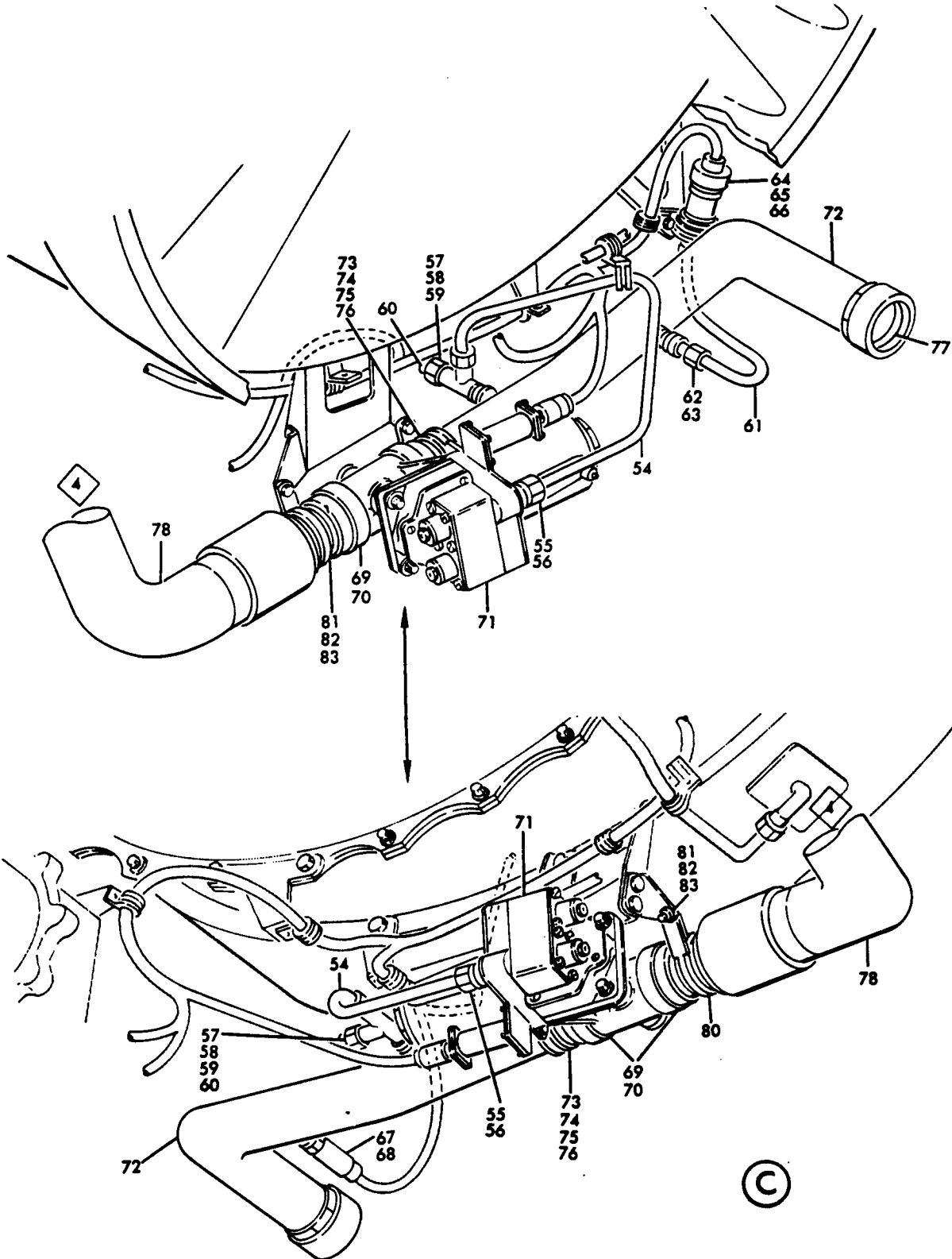




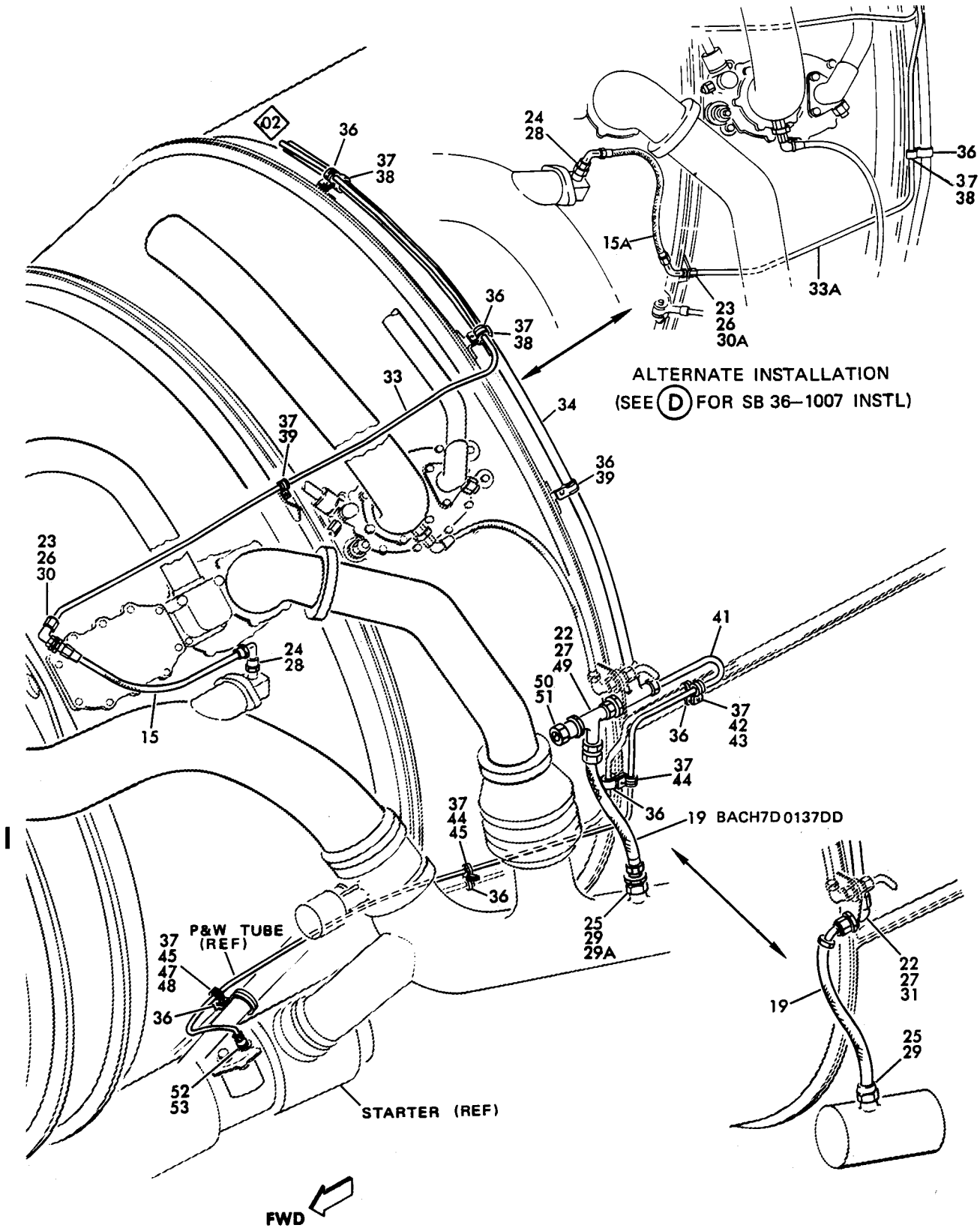
Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 3)



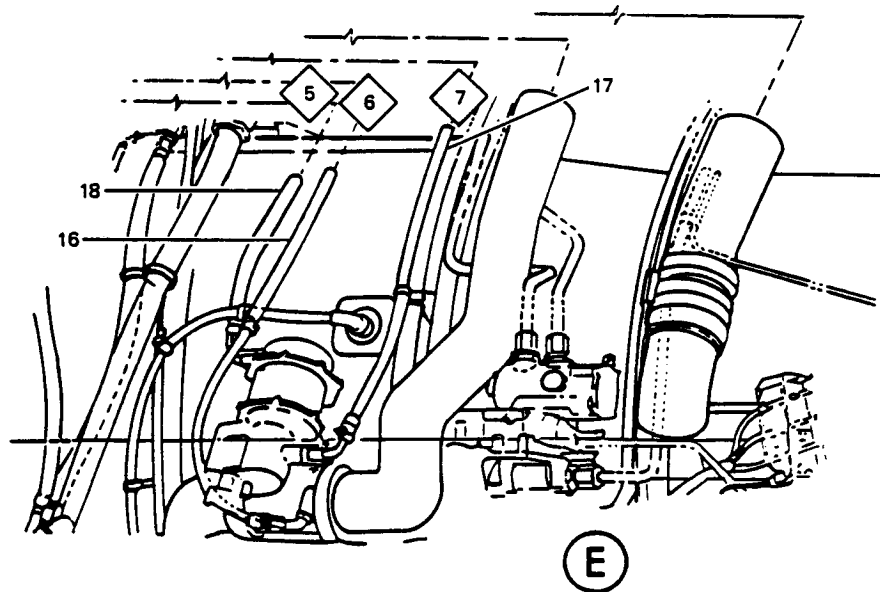
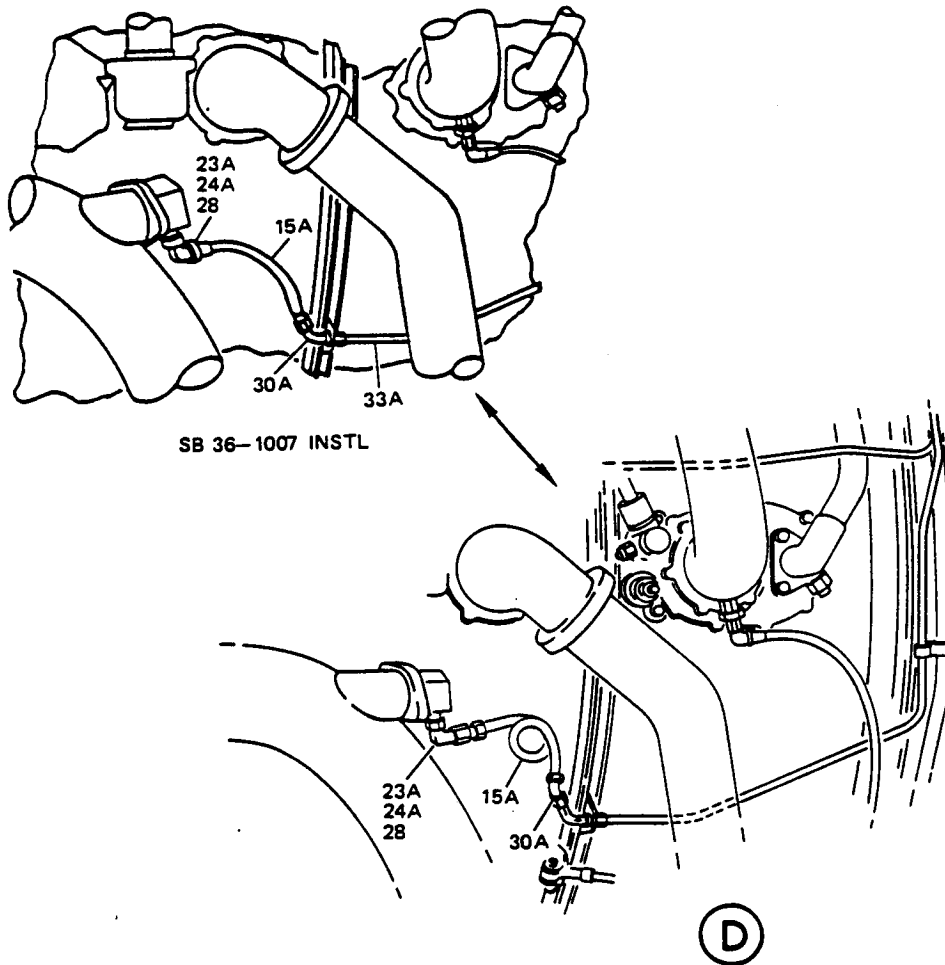
Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 4)



Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 5)



Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 6)



Plumbing Installation, TAI and AC Sensing
Figure 1125 (Sheet 7)

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1125-	65-58126		PLUMBING INSTL, AIR CONDITIONING SENSING AND THERMAL ANTI-ICING								
	65-58126-1		. PLUMBING INSTL, THERMAL ANTI-ICING ICING								
1	BACC10Y22-6C		. . CLAMP *[1]*[7]*[12]								1
1A	NAS1716C20M		. . SADDLE CLAMP *[13]								1
1B	NAS1802-3-8		. . SCREW *[13]								2
2	NAS679A3W		. . NUT *[1]								2
2A	MS21042L3		. . NUT *[13]								2
3	NAS595-20E		. . RING								1
4	BACC10HS20		. . CLAMP *[17]								1
4	BACC10EP20		. . CLAMP *[17]								1
4	BACC10HS20		. . CLAMP *[17]								2
4	BACC10EP20		. . CLAMP *[17]								2
4	BACC10DDC20		. . CLAMP *[1]*[17]								1
4	BACC10DDC20		. . CLAMP *[17]								2
5	BACC10DU125C		. . COUPLING *[1]*[7]								1
6	BACC10DU150C		. . COUPLING *[1]*[7]								1
7	65-58123-10		. . TUBE ASSY *[1]*[7]								1
7	65-58123-1		. . TUBE ASSY *[1]								1
8	65-58123-13		. . TUBE ASSY *[13]								1
8	65-58123-2		. . TUBE ASSY *[1]*[7]*[12]								1
9	65-58123-7		. . TUBE ASSY								1
10	60023-1		. . GASKET, V92764 *[4]								1
11	AN5C10A		. . BOLT *[3]								4
11A	BACB30LM5U10		. . BOLT (REPLS AN5C10A)								2
11B	BACB30LM5U12		. . BOLT (REPLS AN5C10A)								2
12	NAS679C5		. . NUT								4
13	AN960C516		. . WASHER								4
14	BACF22S32AC		. . FLANGE *[1]*[7]								1
	65-58126-2		. PLUMBING INSTL, AIR CONDITIONING SENSING								
15	AS138-04-0084		. HOSE ASSY, V78570 (USE WITH 65-58126-800)(POST SB 36-1007)(PREF)								1
15	16233		. . HOSE ASSY, V81481 (PRE SB 36-1007)								1
15	10643-3		. . HOSE ASSY, V99755 (OPT)								1
15A	65-58126-23		. . HOSE ASSY *[15] (PREF)								1
15A	109339-2		. . HOSE ASSY, V78570 (POST SB 36-1007)								1
15A	AS137-04-0087		. . HOSE ASSY, V78570 (USE WITH 65-58126-21)(SB 21-1032)*[14]								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1125-15A	66081		.	.							1
16	AS136-06-0260		.	.							1
16	16229		.	.							1
16	10648-3		.	.							1
17	AS138-06-0235		.	.							1
17	16228		.	.							1
17	10647-3		.	.							1
18	AS138-08-0134		.	.							1
18	16230		.	.							18
18	10645-3		.	.							1
19	BACH7D0137DD		.	.							1
19	16236		.	.							1
19	10646-3		.	.							1
20	MS21924-6		.	.							1
21	MS21912-6		.	.							1
22	AN924-6		.	.							3
23	AN924-4		.	.							2
23A	AN924-4		.	.							1
24	AN901-4C		.	.							1
24	BACP11PB4		.	.							1
24A	AN901-4C		.	.							1
24A	AN901-4C		.	.							1
25	BACP11PB6		.	.							1
26	AN960-716		.	.							2
27	AN960-916		.	.							3
28	MS21926-4		.	.							1
28	MS21902-4		.	.							1
29	MS21902-6		.	.							1
29A	118-10426		.	.							1
29A	2-01131		.	.							1
30	MS21908-4		.	.							1
30A	MS21907J4		.	.							1
30A	MS21907-4		.	.							1

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1125-30A	BACE21ATO406		.	.	ELBOW (USED WITH 109339-2) (POST SB 36-1007)						1
31	MS21908-6		.	.	ELBOW						1
31	BACT16AXO60406		.	.	TEE, REDUCER (USED WITH 65-58126-12)						1
32	65-58126-801		.	.	TUBE ASSY						
32	65-58126-11		.	.	TUBE ASSY (OPT)						1
33	65-58126-800		.	.	TUBE ASSY (USE WITH 16233)						1
33A	65-58126-21		.	.	TUBE ASSY (USE WITH 66081)(SB 21-1032)						1
34	65-58126-6		.	.	TUBE ASSY						
35	65-58126-10		.	.	TUBE ASSY						
35A	AS136-06-0131		HOSE ASSY, V78570 (USE WITH 65-58126-10)(PREF) *(9) *(10)								1
35A	AS137-06-0146		HOSE ASSY, V78570 (USE WITH 65-58126-10)(POST SB 36-1007) *(9)								1
35A	16235		HOSE ASSY, V81481 (PRE SB 36-1007) *(9)								1
36	BACC10HS06		.	.	CLAMP *[17]						AR
36	BACC10EP6		.	.	CLAMP *[17]						AR
36	BACC10DDC6		.	.	CLAMP *[17]						
37	BACC10HS04		.	.	CLAMP *[17]						AR
37	BACC10EP4		.	.	CLAMP *[17]						AR
37	BACC10DDC4		.	.	CLAMP *[17]						
38	BACS12CB3-9		.	.	SCREW						5
39	BACS12CB3-7		.	.	SCREW *[16]						7
39A	AN960C10		.	.	WASHER						AR
39A	AN960C10L		.	.	WASHER						AR
40	NAS679A3W		.	.	NUT						2
41	65-58126-12		.	.	TUBE ASSY						
42	NAS43HT3-16		.	.	SPACER						1
43	BACS12CB3-12		.	.	SCREW						1
44	BACS12CB3-7		.	.	SCREW						2
45	NAS679A4W		.	.	NUT						2
46	BACC10HS20		.	.	CLAMP *[17]						1
46	BACC10EP20		.	.	CLAMP *[17]						1
46	BACC10DDC20		.	.	CLAMP *[17]						1
47	NAS43HT3-24		.	.	SPACER						1
48	BACS12CB3-16		.	.	SCREW						1
49	118-10430		.	.	TEE, V79470 (USED WITH BACH7D137DD) (SB 75-1002)						1
50	1667		.	.	VALVE, RELIEF, V06177 (USED WITH BACH7D137DD) (SB 75-1002)						1
50	AV18B1037-1		.	.	VALVE, RELIEF, V73760 (SB 75-1102)(OPT)						1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1125-											
51	BACP11PB8										1
52	BACP11PB6										1
53	11-10865										1
53	22103										1
	65-58126-14										
	65-58126-20										
54	65-58126-18										1
54	65-58126-17										1
55	MS21902-4C										1
56	AN901-4C										1
57	MS21912-4C										1
57	MS21912J4										1
58	AN924-4J										1
59	AN901-4C										1
60	BACC14H4LJ										1
61	65-58126-19										1
62	MS21902-4J										1
63	AN901-4C										1
	65-77451-2										
	65-77451-1										
	65-77451-3										1
	65-77451-4										
64	7026										1
64	7026-1										1
65	BACC10HS18										1
65	BACC10EP18										1
66	BACS12CB3-7										1
67	8G654										1
68	AN901-4C										1
69	24502-150										2
70	24096-150C										2

737

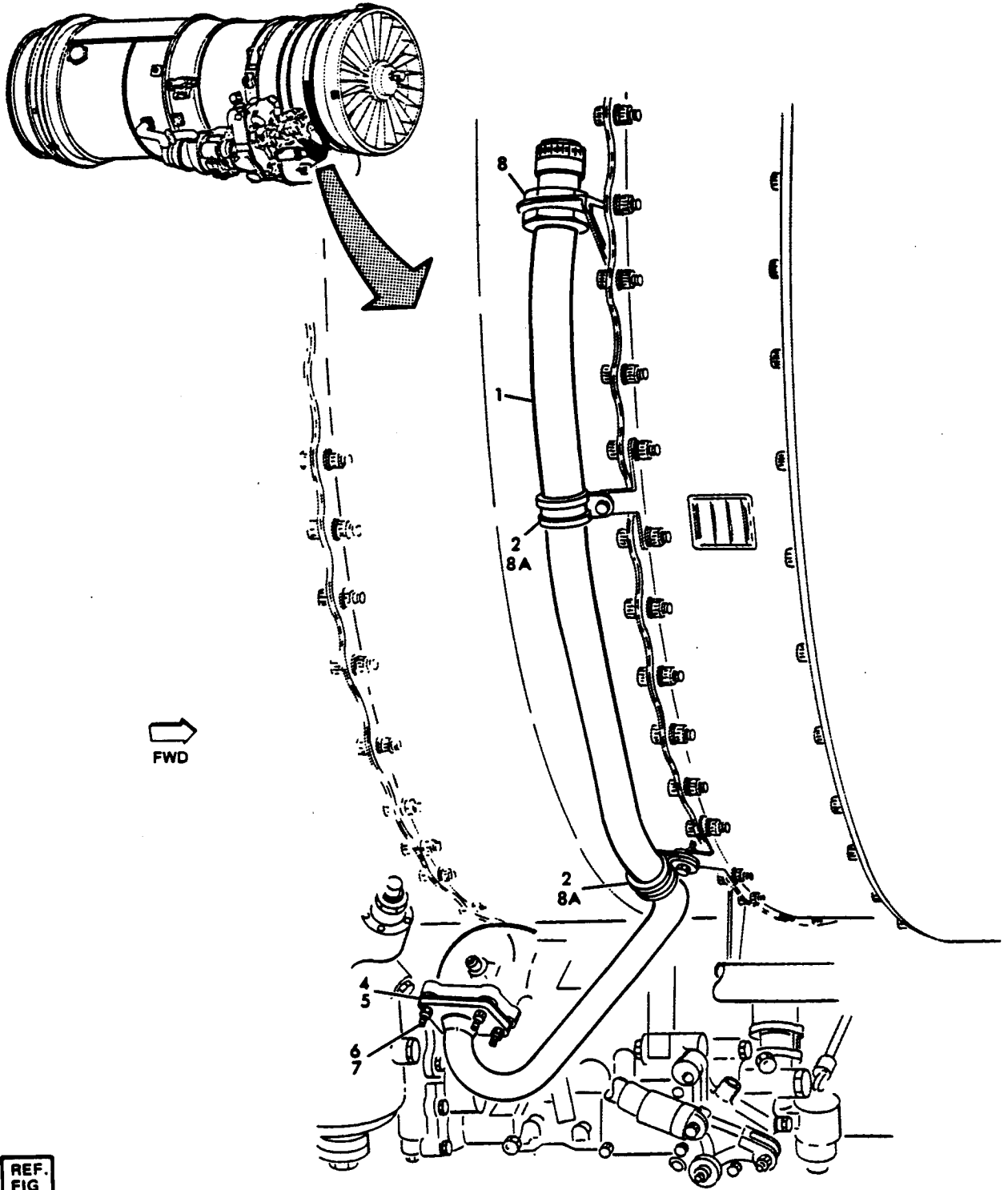
POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1125-71	F61D0236M4		.	.	.	VALVE, V92003 (USED ON 65-77451-2, -3)						1
71	F61D0236M3		.	.	.	VALVE, V92003 (USED ON 65-77451-1)						1
71	F61D0236M6		.	.	.	VALVE V92003 (USED ON 65-77451-4)(SB 75-1004) *[8]						1
72	65-77452-10		.	.		DUCT ASSY						1
72	65-77452-16		.	.		DUCT ASSY *[8]						1
73	BACC10HS24		.	.		CLAMP *[17]						1
73	BACC10EP24		.	.		CLAMP *[8]*[17]						1
74	BACS12CB3-6		.	.		SCREW						1
75	BACN10JC3		.	.		NUT						1
76	AN960-10L		.	.		WASHER						1
77	NAS595-24E		.	.		RING						1
78	65-77452-1		.	.		DUCT ASSY						1
78	65-77452-13		.	.		DUCT ASSY (PREF ON 65-58126-20) *[8]						1
79	BACC10DU200X		.	.		COUPLING *[8]						1
80	BACC10HS24		.	.		CLAMP *[17]						1
80	BACC10EP24		.	.		CLAMP *[17]						1
81	BACS12CB3-6		.	.		SCREW						1
82	BACN10JC3		.	.		NUT						1
83	AN960-10L		.	.		WASHER						1
84	BACC10HS32		.	.		CLAMP *[17]						1
84	BACC10EP32		.	.		CLAMP *[17]						1
85	BACC10HS06		.	.		CLAMP *[17]						1
85	BACC10EP6		.	.		CLAMP *[17]						1
86	BACS12CB3-6		.	.		SCREW						1
87	BACN10JC3		.	.		NUT						1
88	NAS43HT3-16		.	.		SPACER						1
89	S387SSM32		.	.		CLAMP (PREF), V18076						2
89	BACC10CD32		.	.		CLAMP						2
90	BACS12CB3-6		.	.		SCREW						4
91	BACN10JC3		.	.		NUT						4
92	65-58126-24		.			PLUMBING INSTL, STARTER VALVE OPEN SWITCH						1
92	65-58126-25		.			PLUMBING INSTL, STARTER VALVE OPEN SWITCH						1
93	65-58126-26		.	.		TUBE ASSY (USED ON 65-58126-24) *[11]						1
93	65-58126-27		.	.		TUBE ASSY (USED ON 65-58126-25) *[11]						1

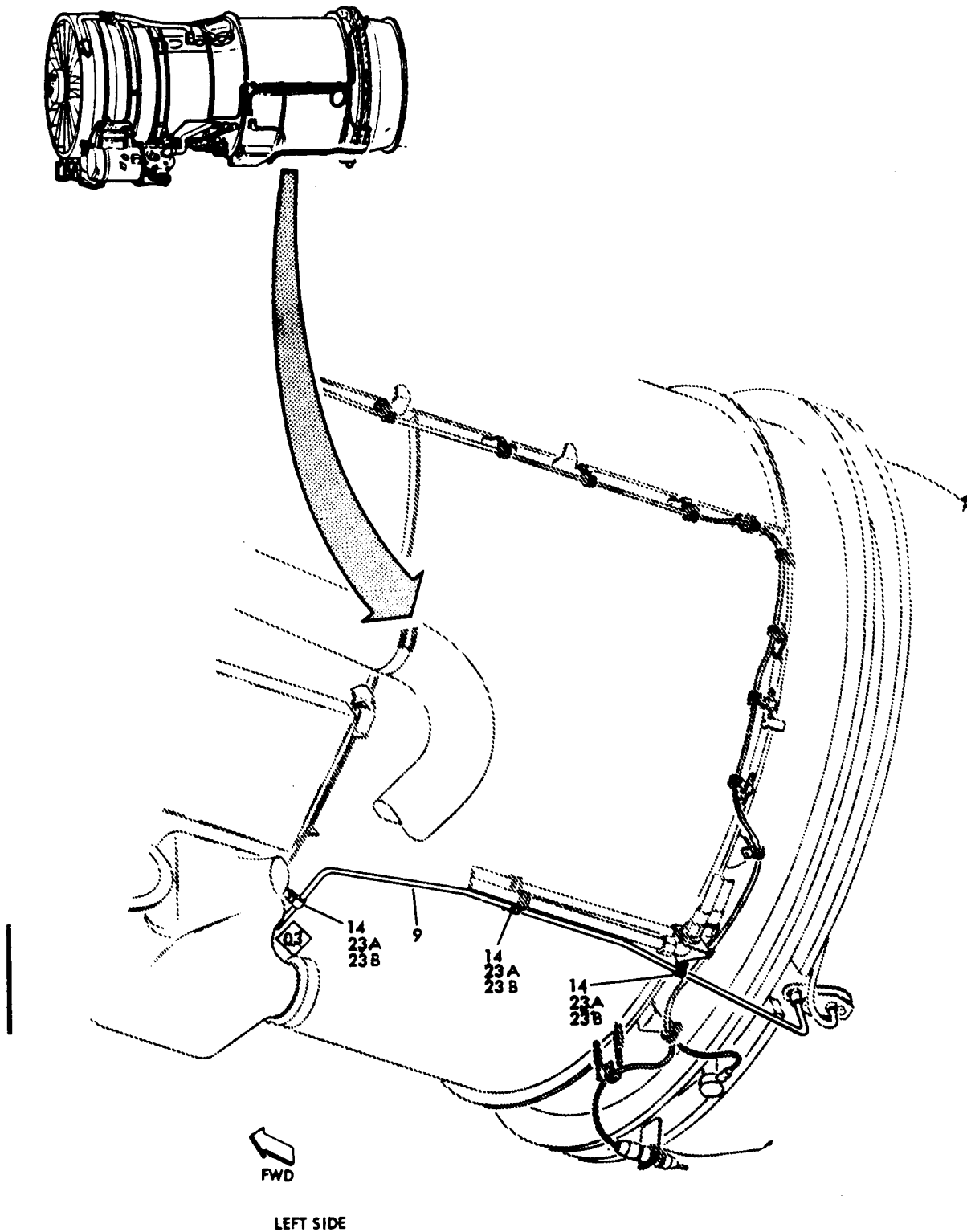
- *[1] NOT REQUIRED ON POWER PLANT ASSEMBLIES WITH BOOM TYPE VORTEX DISSIPATOR
- *[2] INSTALLATION COVERED BY KIT DRAWING 65-75985
- *[3] OPTIONAL TO BACB30LM5010 AND BACB30LM5012
- *[4] REPLACES 60023
- *[5] USED ON 65-58126-20
- *[6] OPTIONAL ON 65-58126-14 PER SB 71-1015
- *[7] PRE SB 71-1102, 71-1111
- *[8] POST SB 71-1102, 71-1111
- *[9] HOSE ASSY NOT PART OF QEC BUT IS INCLUDED TO SHOW STRUT CONNECTION
- *[10] PREFERRED HOSE ASSY ON ENGINE No. 1 WHEN POTABLE WATER PRESSURE INSTALLATION IS USED.
- *[11] SEE FIG. 1103 FOR DETAILS
- *[12] PRE SB 30-1029
- *[13] POST SB 30-1029
- *[14] PRE SB 36-1007
- *[15] POST SB 36-1007
- *[16] INCREASE IN SCREW LENGTH MAY BE NECESSARY TO ACCOMMODATE SHIM (39A) INSTALLATION
- *[17] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC() CLAMPS



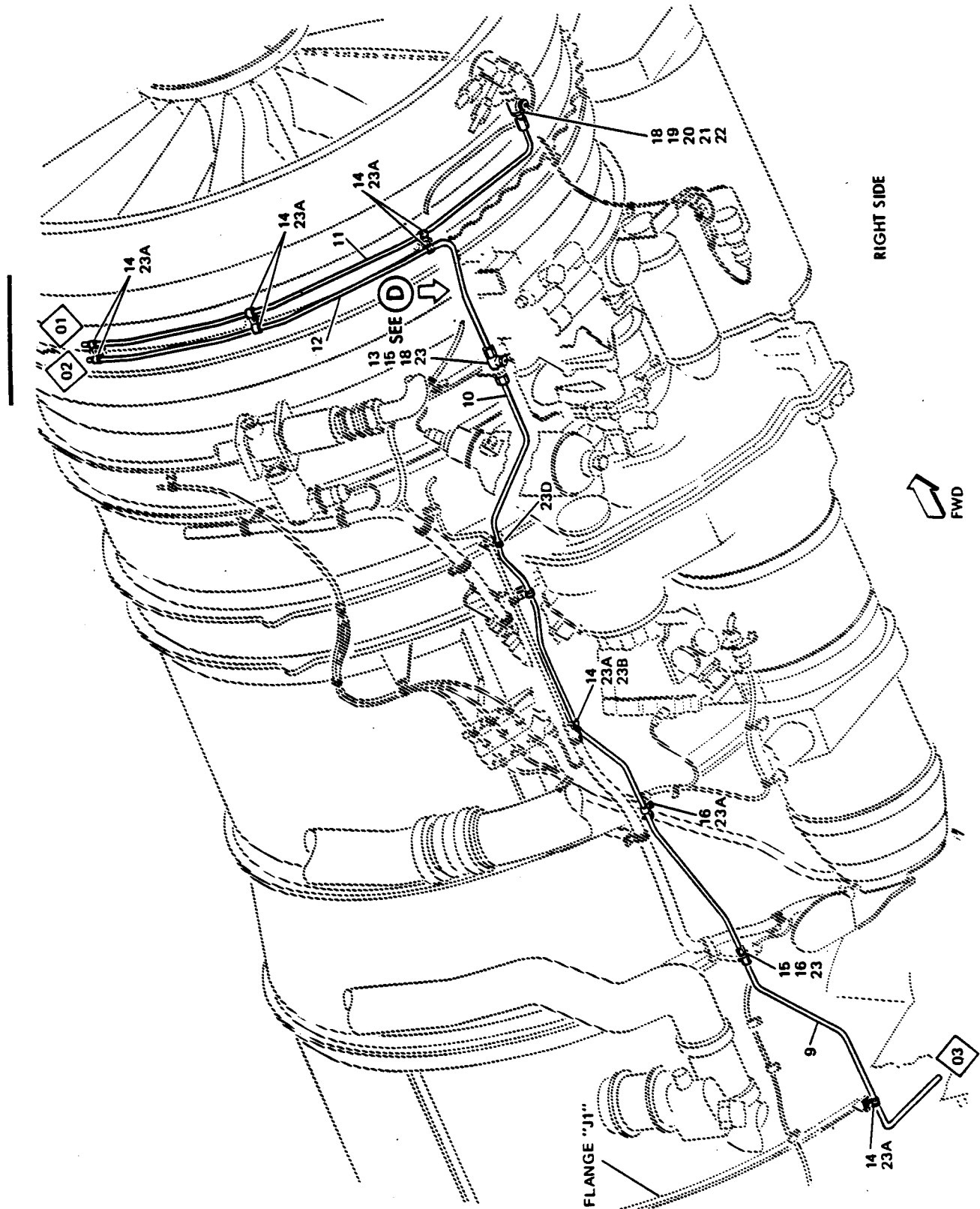
RIGHT SIDE

REF.
FIG.
524
535
543
545

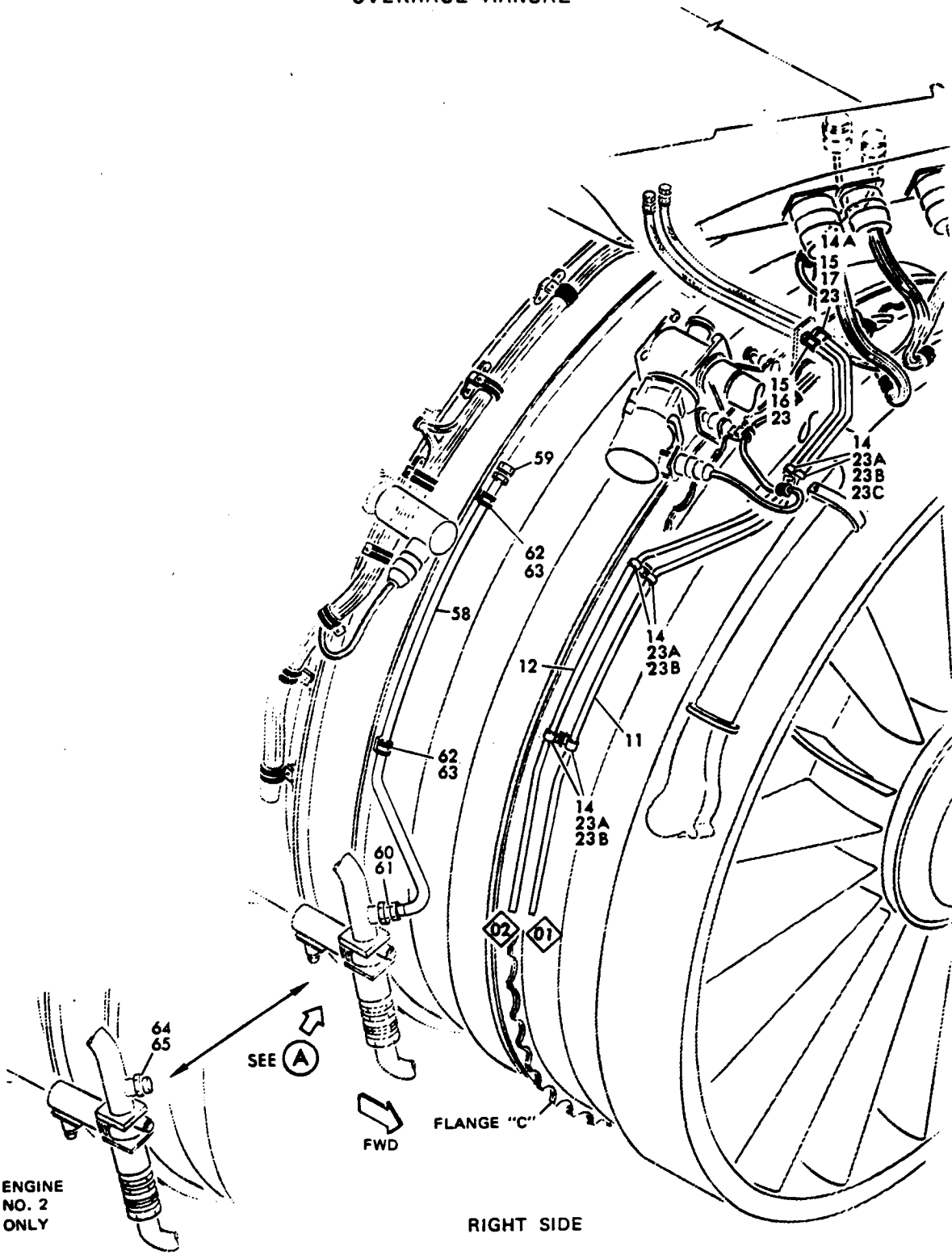
Plumbing Installation, Engine Systems
Figure 1126 (Sheet 1)



Plumbing Installation, Engine Systems
Figure 1126 (Sheet 2)

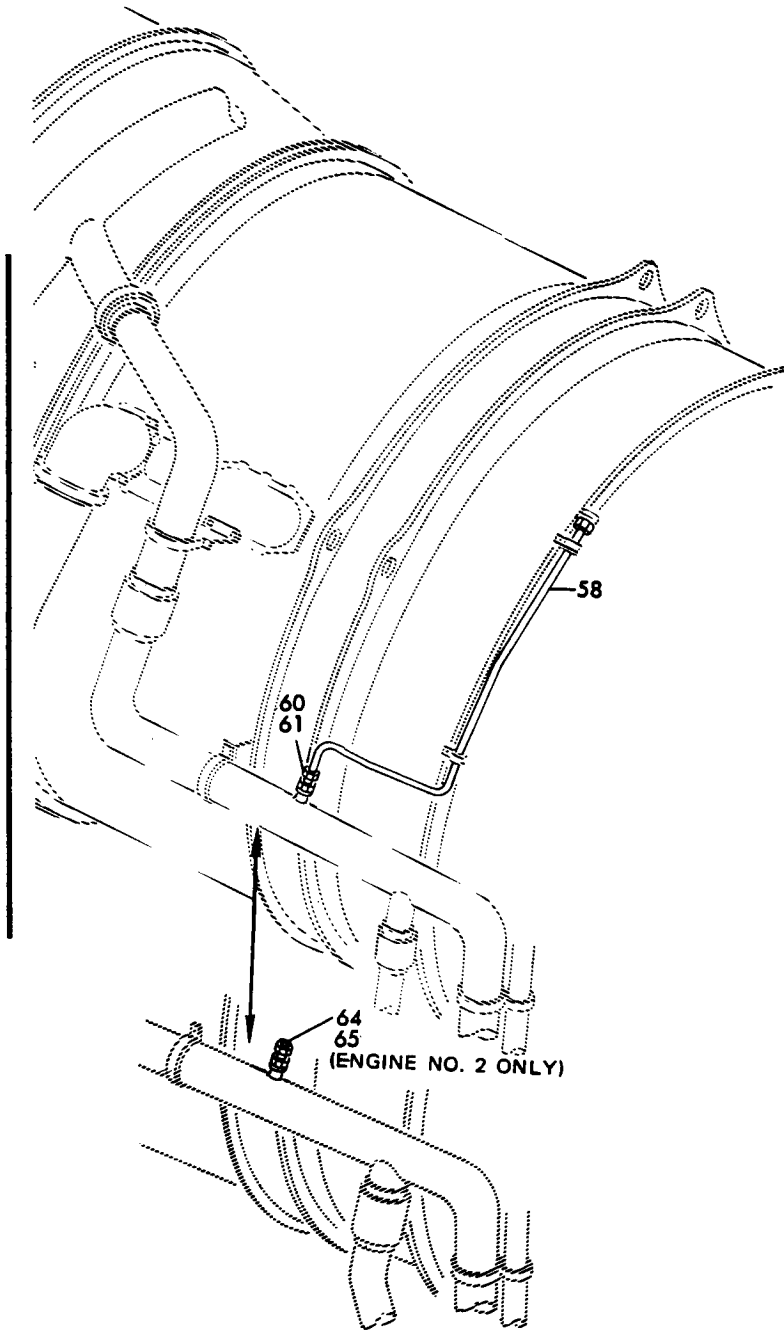


Plumbing Installation, Engine Systems
Figure 1126 (Sheet 3)



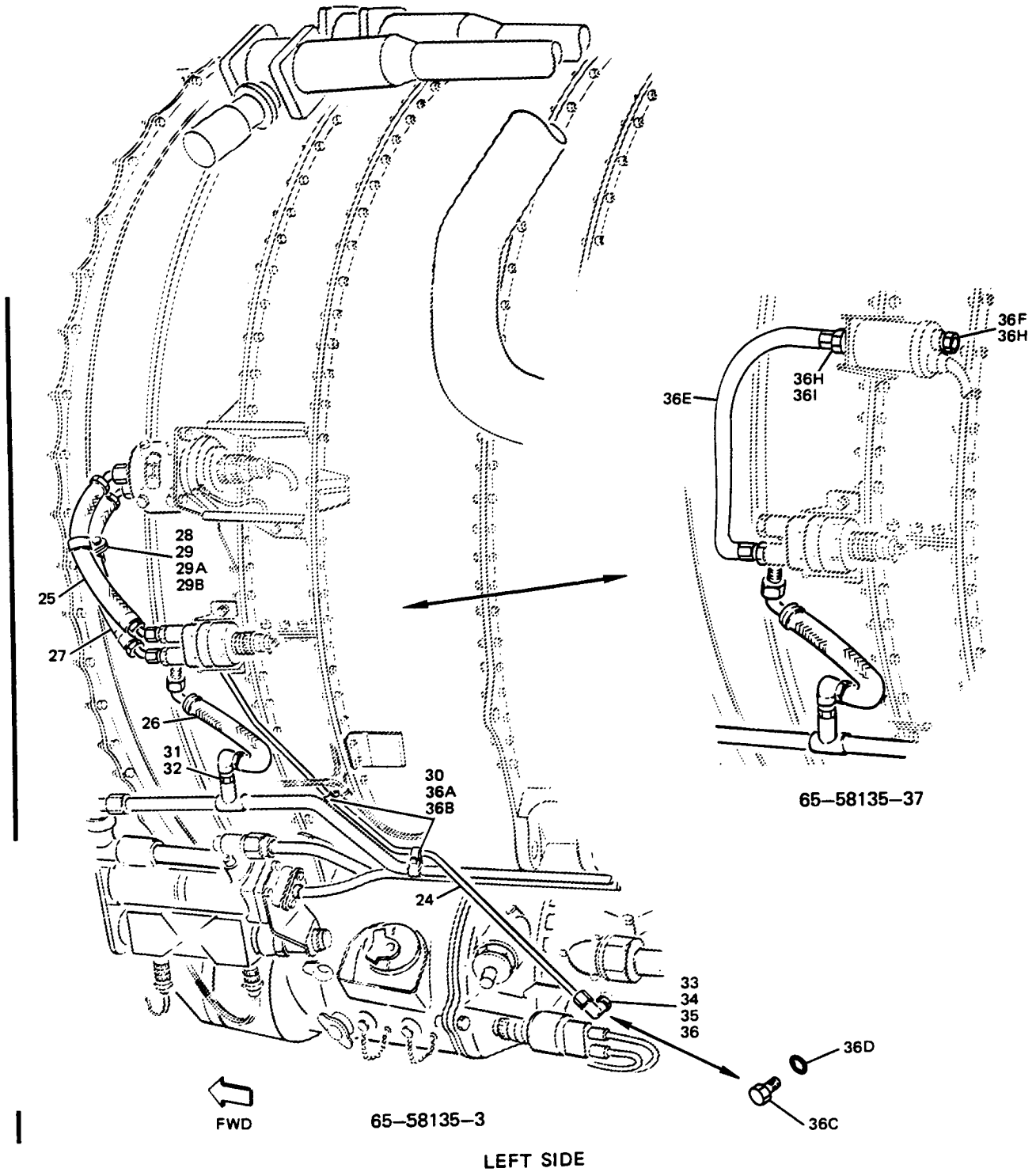
Plumbing Installation, Engine Systems
Figure 1126 (Sheet 4)

Jan 5/83

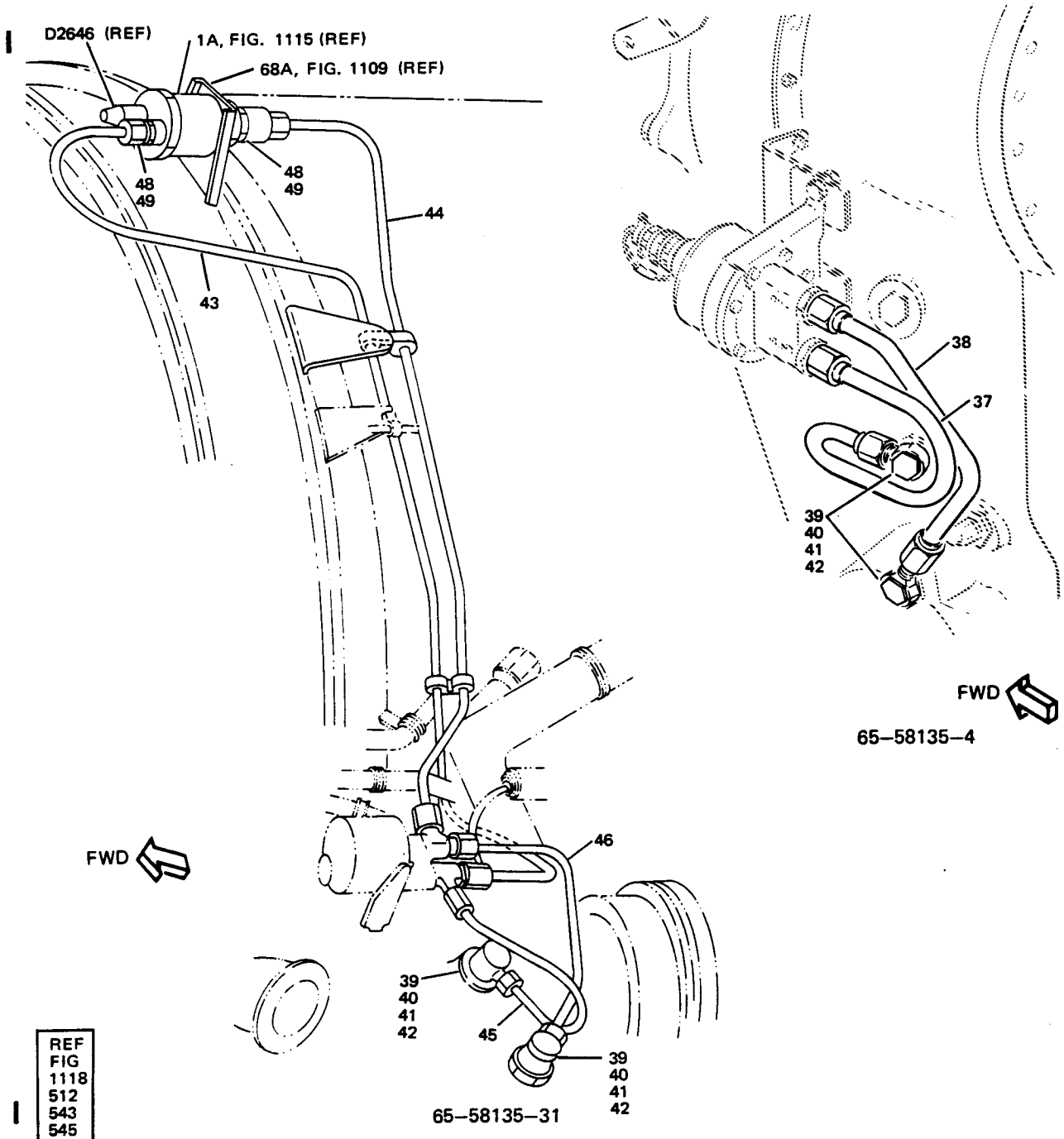
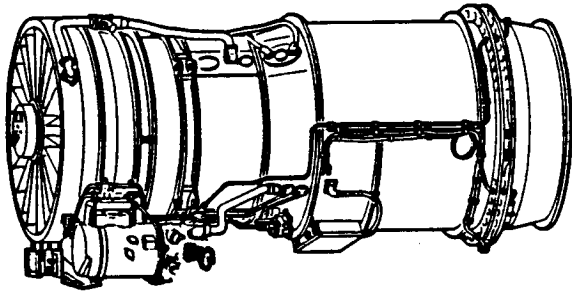


(A)

ENGINES WITH VORTEX DISSIPATOR

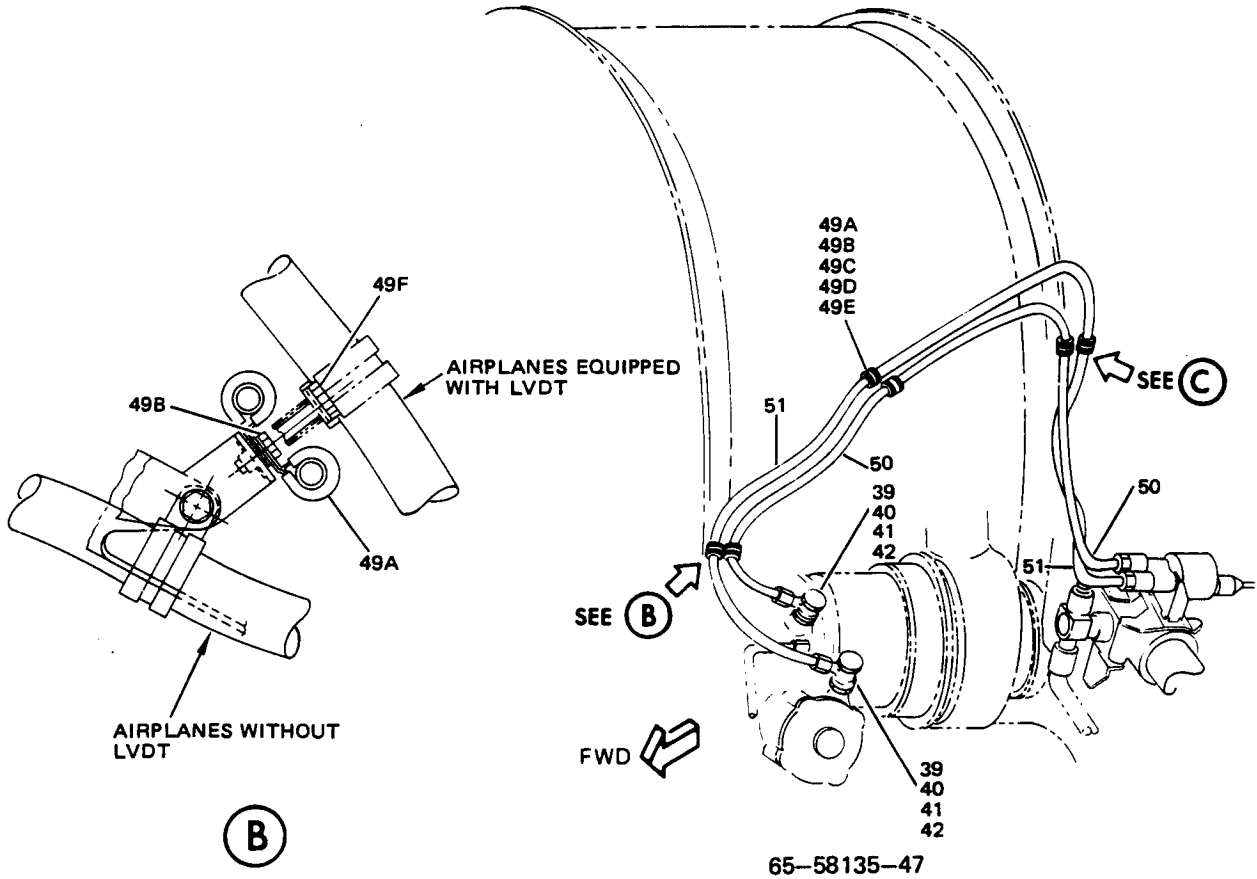
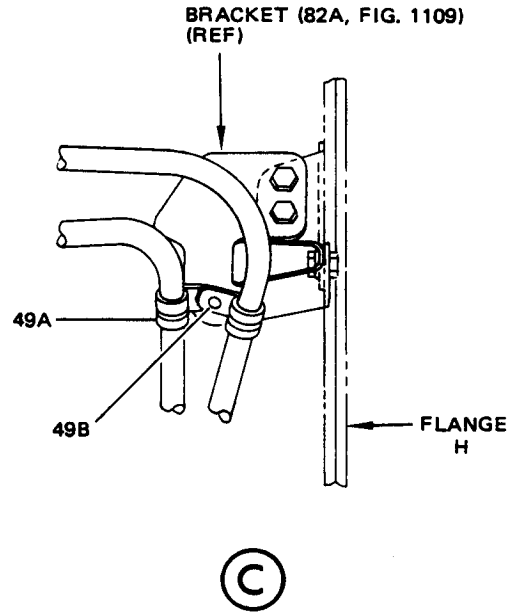
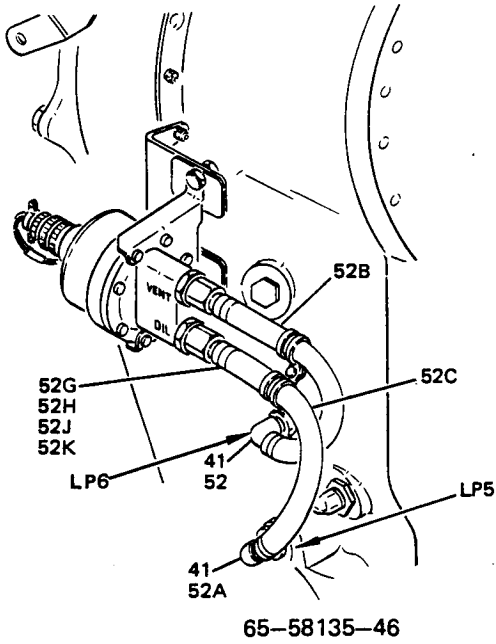


Plumbing Installation, Engine Systems
Figure 1126 (Sheet 6)

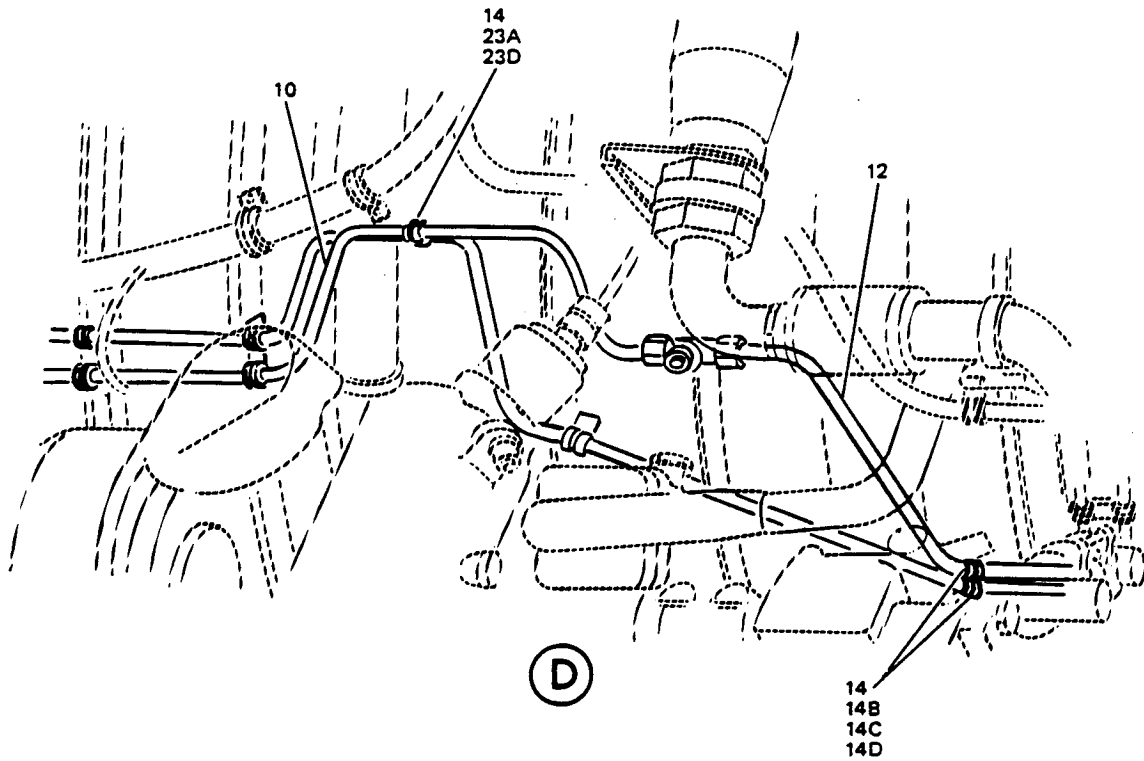


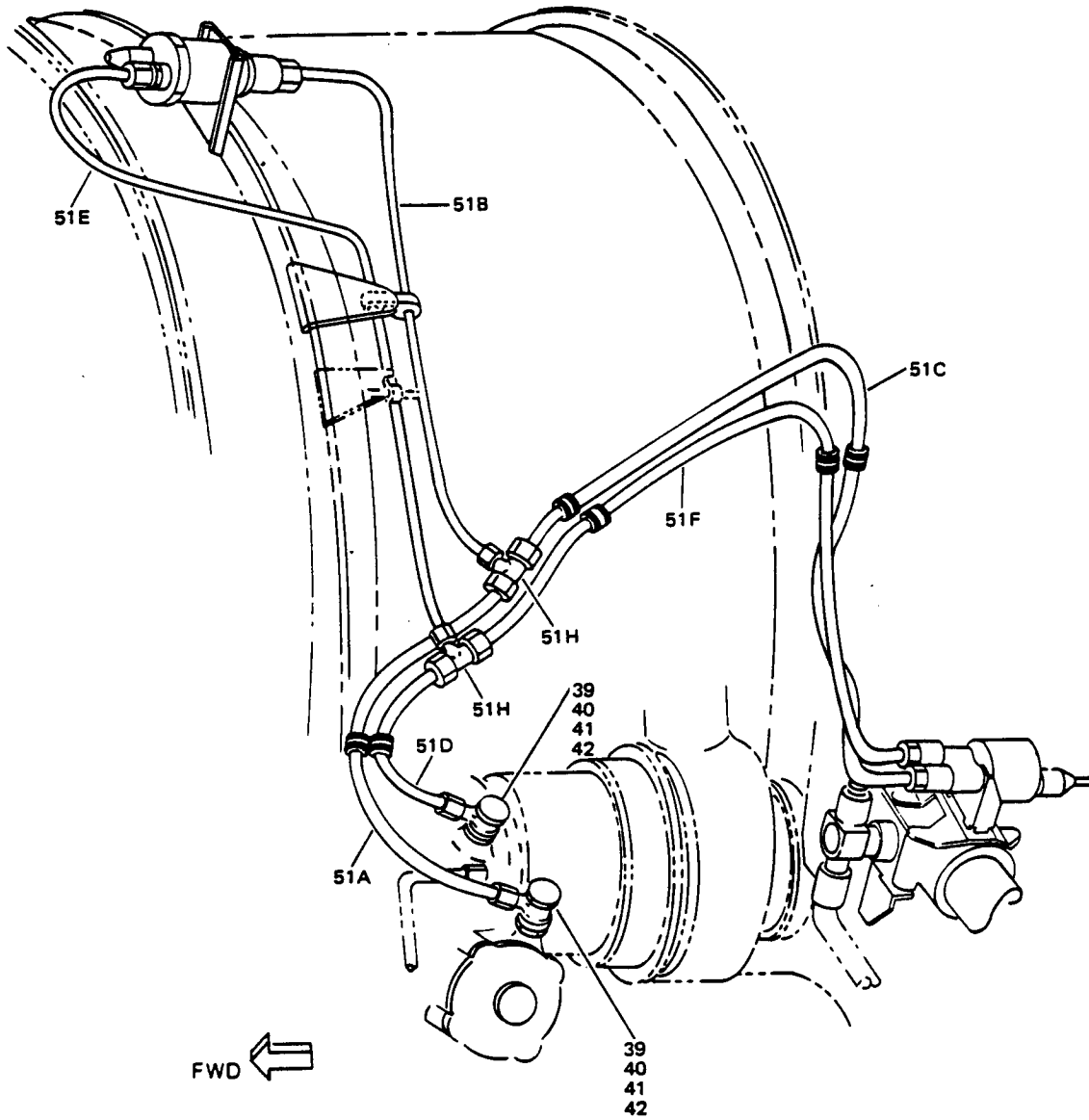
REF
FIG
1118
512
543
545

Plumbing Installation Engine Systems
Figure 1126 (Sheet 7)



Plumbing Installation Engine Systems
Figure 1126 (Sheet 8)





65-58135-48

737

POWERPLANT BUILDUP

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1126-	65-58135		PLUMBING INSTL ENGINE SYSTEMS								
	65-58135-1		. PLUMBING INSTL, FUEL SUPPLY								RF
1	65-46869-4		. . TUBE ASSY								1
1	65-46869-800		. . TUBE ASSY (OPT)								1
2	BACC10HS16		. . CLAMP *[8]								2
2	BACC10EP16		. . CLAMP *[8]								2
2	BACC10DDC16		. . CLAMP *[8]								2
3			. . DELETED								
4	MS20756-24		. . FLANGE								1
5	BACG10AG305		. . GASKET								1
6	AN320-4		. . NUT								4
7	AN960PD416L		. . WASHER								4
8	AN924-16		. . NUT								1
8A	BACS12CB3-7		. . SCREW								2
8B	BACC10HS23		. . CLAMP *[8]								1
8B	BACC10EP23		. . CLAMP *[8]								1
8B	BACC10DDC23		. . CLAMP *[8]								1
	65-58135-2		. PLUMBING INSTL, PRESSURE RATIO								RF
	65-58135-26		. PLUMBING INSTL, PRESSURE RATIO (JT8D-17)								RF
	65-58135-30		. PLUMBING INSTL, PRESSURE RATIO (JT8D-17, ACCOMMODATES NEW P & W FUEL PRESS. DIFF. SWITCH)								RF
9	65-58135-5		. . TUBE ASSY								1
10	65-58135-6		. . TUBE ASSY (USED ON 65-58135-2) *[5] *[10] *[11]								1
-10	65-58135-28		. . TUBE ASSY (SB 71-1034, -1041)								1
10	65-58135-29		. . TUBE ASSY (USED ON 65-58135-30) *[6] *[9] *[12]								1
11	65-58135-8		. . TUBE ASSY								1
12	65-58135-13		. . TUBE ASSY (USED ON 65-58135-2) *[5] *[10] *[11]								1
-12	65-58135-27		. . TUBE ASSY (SB 71-1041, -1058) *[6] *[9] *[12]								1
13	MS21912-4		. . TEE								1
13	MS21912-4J		. . TEE (OPT)								1
-13A	BACS12CB37		. . SCREW (SB 71-1041)								1
-14	BACC10HS04		. . CLAMP *[8]								1
-14	BACC10EP4		. . CLAMP (SB 71-1041) *[8]								1
-14	BACC10HS05		. . CLAMP *[8]								1
-14	BACC10EP5		. . CLAMP (SB 71-1041) *[8]								1
14	BACC10HS04		. . CLAMP *[8]								20
14	BACC10EP4		. . CLAMP *[7] *[8]								20
14	BACC10DDC4		. . CLAMP *[8]								20

- ITEM NOT ILLUSTRATED

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1126-14A	AN924-6										1
14A	AN924-4										1
14B	NAS1801-3-10										2
14C	AN960C11										2
14D	NAS679C3M										2
15	AN924-4										3
16	MS21924-4										2
16	MS21924-4J										2
17	2-01071										1
17	GA7850										1
17	BACU24AA0406										1
18	BACC14H4L										2
19	NAS1237-4										1
20	NAS1236-4H										1
21	BACP11PB4										1
22	BACP11P11										1
23	AN960-716										4
23	AN960-916										1
23A	BACSL2CB3-(-)										AR
23B	BACN10JC3										AR
23C	BACSL8AN12										1
23D	NAS43HT3-48. 65-58135-3										1 RF
	65-58135-37										RF
24	65-58135-14										1
25	10-61357-2										1
26	10-61357-3										1

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1126-27	10-61357-4		. . HOSE ASSY (USED ON 65-58135-3) (PRE SB 79-1003) *[2] B520DDB0104-000, V98441 AE702087-2, V70128 S41B90041060105, V79470								1
28	BACC10HS12		. . CLAMP *[8] (USED ON 65-58135-3) (PRE SB 79-1003) *[2]								1
28	BACC10EP12		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
28	BACC10DDC12		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
29	BACC10HS10		. . CLAMP *[8] (USED ON 65-58135-3) (PRE SB 79-1003) *[2]								1
29	BACC10EP10		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
29	BACC10DDC10		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
29A	BACS12CB3-7		. . SCREW *[1] (USED ON 65-58135-3) (PRE SB 79-1003) *[2]								1
29B	BACN10JC3		. . NUT (REPL NAS679A3) *[1] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
30	BACC10HS04		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								2
30	BACC10EP4		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								2
30	BACC10DDC4		. . CLAMP *[1]*[8] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								2
31	D9798		. . UNION, V14798 *[2]								1
31	1-01288		. . UNION, V11328 *[2](OPT TO D9798)								1
31	69-64545-1		. . UNION, (OPT TO D9798) *[2]								1
31	69-64545-4		. . UNION, (REWORK OF 118-10398 or 1-0118)(OPT TO 69-64545-1)*[2]								1
31	1-01188		. . UNION, V11328 (REPLACE OR REWORK) *[2]								1
31	118-10398		. . UNION, RESTRICTOR, (V79470 OR V92003)(OPT) *[2]								1
32	BACP11PB6		. . PACKING, O-RING								1
33	MS21908-4		. . ELBOW *[1] (USED ON 65-58135-3) (PRE SB 79-1003) *[2]								1
34	AN6289-4		. . NUT *[1] (USED ON 65-58135-3) (PRE SB 79-1003) *[2]								1

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1126-35	BACP11PB4		. . PACKING, O-RING *[1] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
36	MS28777-4		. . RING, BACKUP *[1] (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								1
36A	BACS12CB3-7		. . SCREW (USED ON 65-58135-3)(PRE SB 79-1003) *[2]								2
36B	NAS679A3W		. . NUT (REPLD BY BACN10JC3)(USED ON 65-58135-3)(PRE SB 79-1003) *[2]								2
36B	BACN10JC3		. . NUT (REPLS NAS679A3W)								2
36C	MS9015-04		. . PLUG (POST SB 79-1003)								1
36D	MS9387-04		. . PACKING, O-RING (POST SB 79-1003)								1
36E	65-58135-36		. . TUBE ASSY (USED ON 65-58135-37)								1
36F	66-13955-875		. . VENT PORT (USED ON 65-58135-37)								1
36H	BACP11PB4		. . PACKING (USED ON 65-58135-37)								2
36I	MS21916J6-4		. . REDUCER (USED ON 65-58135-37)								1
	65-58135-4		. PLUMBING INSTL, OIL FILTER DIFFERENTIAL PRESSURE SWITCH								RF
	65-58135-31		. PLUMBING INSTL, OIL FILTER DIFFERENTIAL PRESSURE SWITCH								RF
	65-58135-46		. PLUMBING INSTL, OIL FILTER DIFFERENTIAL PRESSURE SWITCH								RF
	65-58135-47		. PLUMBING INSTL, OIL FILTER DIFFERENTIAL PRESSURE SWITCH								RF
	65-58135-48		. PLUMBING INSTL, OIL FILTER DIFFERENTIAL PRESSURE SWITCH								RF
37	65-58135-15		. . TUBE ASSY (USED ON 65-58135-4) (PRE SB 79-1004)								1
38	65-58135-16		. . TUBE ASSY (USED ON 65-58135-4) (PRE SB 79-1004)								1
39	2-01575		. . ELBOW, RESTRICTOR, (V11328)								2
39	118-10557		. . ELBOW, RESTRICTOR, (V79470 OR V92003)(OPT)								2
40	NAS551-4H		. . BOLT, UNIVERSAL								2
41	BACP11PB4		. . PACKING, O-RING (POST SB 79-1004)								2
42	BACP11P11		. . PACKING, O-RING								2
43	65-58135-32		. . TUBE ASSY (USED ON 65-58135-31)								1
44	65-58135-33		. . TUBE ASSY (USED ON 65-58135-31)								1
45	65-58135-34		. . TUBE ASSY (USED ON 65-58135-31)								1
46	65-58135-35		. . TUBE ASSY (USED ON 65-58135-31)								1
47	66-13955-875		DELETED								
48	BACP11PB4		. . O-RING (USED ON 65-58135-31)								2
49	MS21916J6-4		. . REDUCER (USED ON 65-58135-31)								2

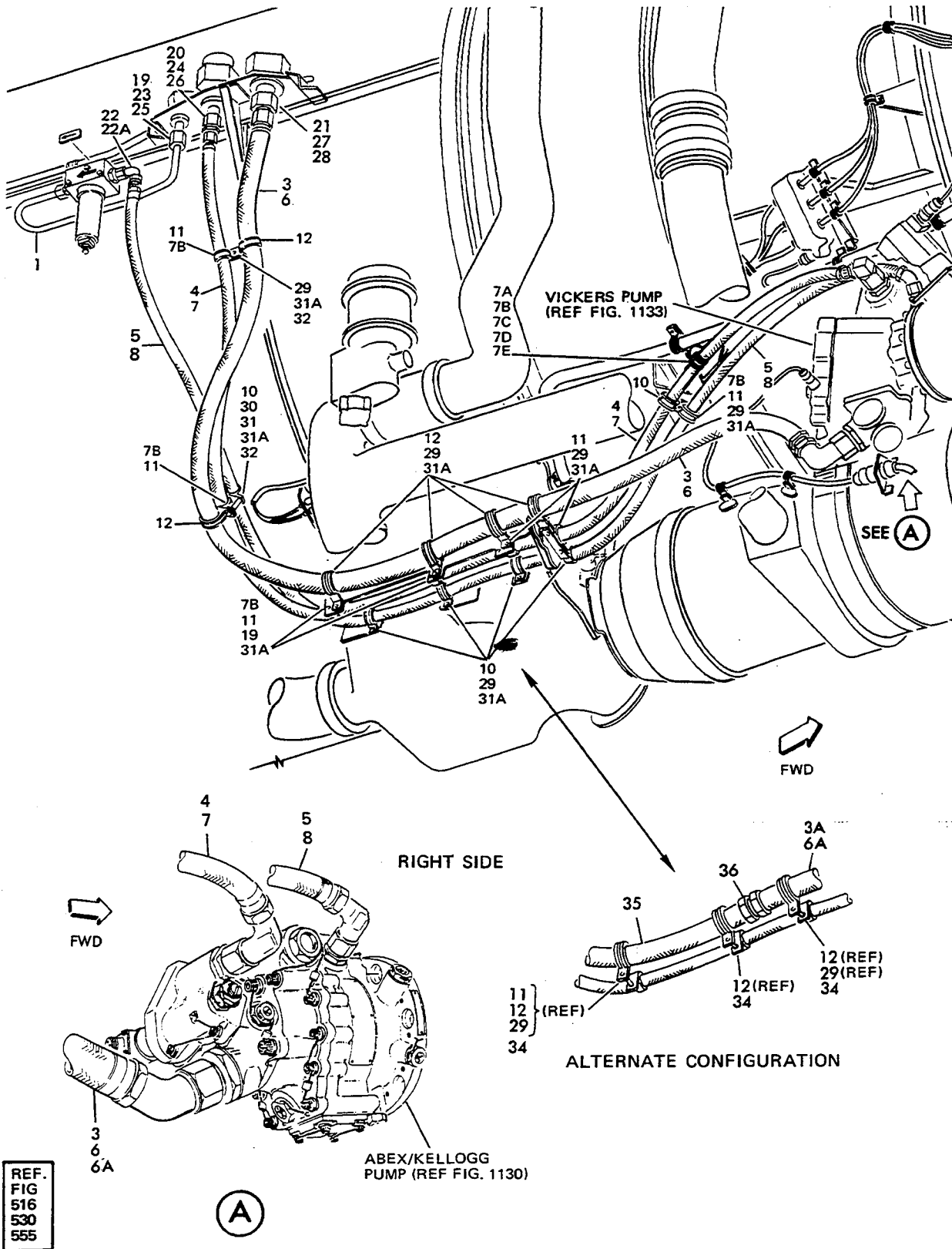
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1126-49A	BACC10HS06		.	.							3
49A	BACC10EP6		.	.							3
49B	NAS1801-3-9		.	.							3
49C	BACC10HS20		.	.							1
49C	BACC10EP20		.	.							1
49D	AN960D10L		.	.							1
49E	BACN10JC3		.	.							1
49F	NAS1801-3-22		.	.						AR	
50	65-58135-38		.	.							1
51	65-58135-39		.	.							1
51A	65-58135-40		.	.							1
51B	65-58135-41		.	.							1
51C	65-58135-42		.	.							1
51D	65-58135-43		.	.							1
51E	65-58135-44		.	.							1
51F	65-58135-45		.	.							1
51G	BACP11PB4		.	.							2
51H	MS21905-6		.	.							2
51J	MS21916J6-4		.	.							2
52	69-72689-1		.	.							1
52A	69-72689-2		.	.							1
52B	AE3660121E0066		.	.							1
52B	SR50206CC4-0066		.	.							1

737

POWERPLANT BUILDUP

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1126-52C	AE3660131E0082		.	.						HOSE ASSY (USED ON 65-58135-46) (POST SB 79-1004)		1
52C	SR50306CC4-0082		.	.						HOSE ASSY (USED ON 65-58135-46) (OPT) (POST SB 79-1004)		1
52G	BACC10HS11		.	.						CLAMP *[8]		2
52G	BACC10EP-11		.	.						CLAMP (POST SB 79-1004) *[8]		2
52H	BACS12CB3-8		.	.						SCREW (POST SB 79-1004)		1
52J	AN960D10L		.	.						WASHER (POST SB 79-1004)		1
52K	BACN10JC3		.	.						NUT (POST SB 79-1004)		1
	65-58135-18		.	.						PLUMBING INSTL, POTABLE WATER PRESSURE		RF
	65-58135-21		.	.						PLUMBING INSTL, POTABLE WATER PRESSURE		RF
58	65-58135-19		.	.						TUBE ASSY (USED ON 65-58135-18) *[3]		1
58	65-58135-25		.	.						TUBE ASSY (USED ON 65-58135-21) (SB 78-1005) *[4]		1
59	MS21902-6J		.	.						UNION (USED ON 65-58135-18)		1
60	118-10530		.	.						FITTING, SPECIAL (V88334 OR V92003)(USED ON 65-58135-18) *[4]		1
61	AN901-8C		.	.						GASKET (USED ON 65-58135-18)		1
62	BACC10HS06		.	.						CLAMP *[8]		2
62	BACC10EP6		.	.						CLAMP *[8]		2
63	BACS12CB3-7		.	.						SCREW		2
	65-58135-20		.	.						PLUMBING INSTL, PLUG		RF
64	BACP20AU8J		.	.						PLUG		1
65	AN901-8C		.	.						GASKET		1

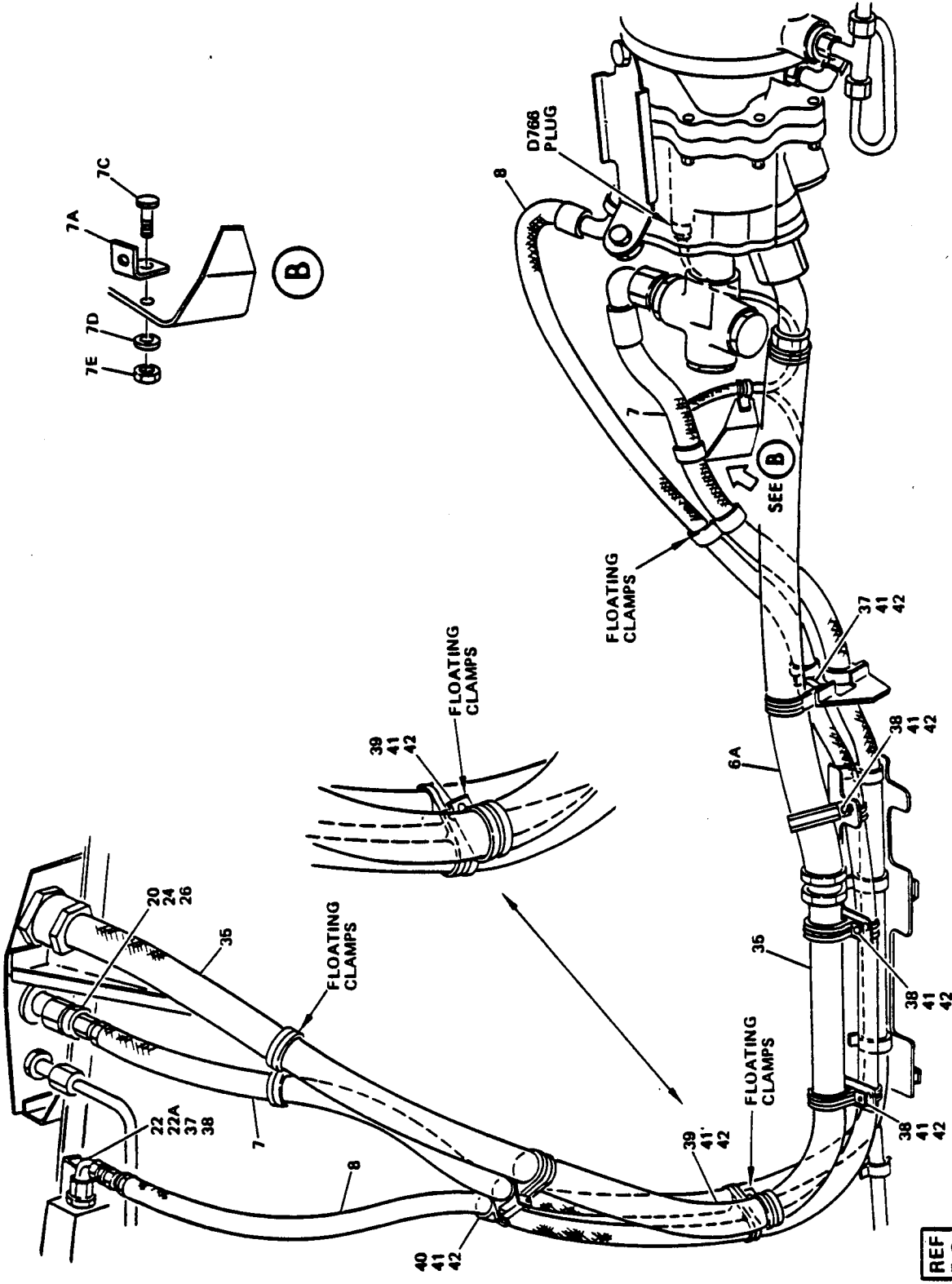
- *[1] REQUIRED ONLY ON SYSTEMS HAVING OIL PRESSURE SWITCH AND TRANSMITTER VENTED TO ENGINE GEARBOX.
- *[2] LIMITED USAGE.
- *[3] PRE SB 71-1102, 71-1111
- *[4] POST SB 71-1102, 71-1111
- *[5] PRE SB 71-1108, -1099, -1078, -1104, -1138, -1150, -1155, -1154, -1159, -1180, -1182, -1192, -1255, -1263, -1337
- *[6] POST SB 71-1108, -1099, -1078, -1104, -1138, -1150, -1155, -1154, -1159, -1180, -1182, -1192, -1255, -1263, -1337
- *[7] POST SB 71-1108, -1182, -1192
- *[8] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC()
CLAMPS
- *[9] PRE SB 71-1340
- *[10] POST SB 71-1340
- *[11] PRE SB 71-1351
- *[12] POST SB 71-1351



REF.
FIG
516
530
555

(A)

Plumbing Installation, Hydraulic System
Figure 1127 (Sheet 1)

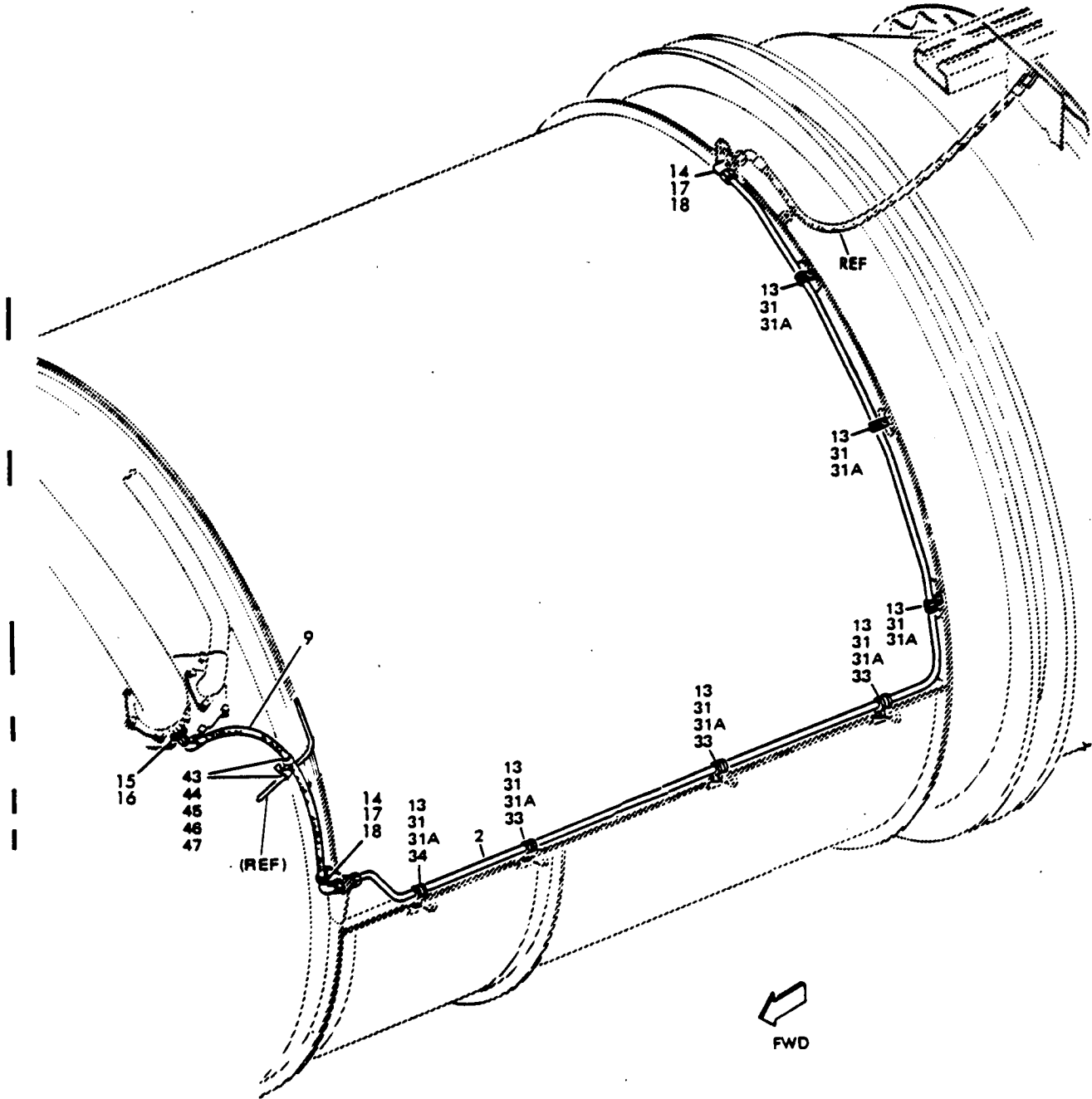
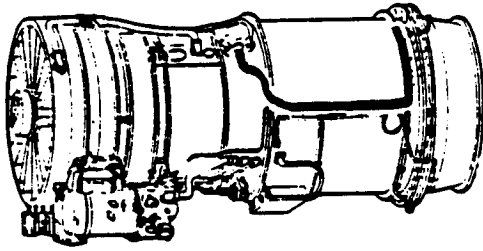


VICKERS IN-LINE TYPE HYDRAULIC PUMP PLUMBING INSTALLATION

65-58145-15

REF	FIG
516	530
555	

OVERHAUL MANUAL



Plumbing Installation, Hydraulic System
Figure 1127 (Sheet 3)

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1127-	65-58145-1		PLUMBING INSTL (VICKERS PUMP)							A	
	65-58145-4		PLUMBING INSTL (ABEX PUMP)							B	
	65-58145-9		PLUMBING INSTL (ABEX PUMP WITH QAD)							C	
	65-58145-10		PLUMBING INSTL (KELLOGG PUMP)							D	
	65-58145-12		PLUMBING INSTL (VICKERS PUMP)							E	
	65-58145-14		PLUMBING INSTL, RIPPLE ATTENUATOR							F	
	65-58145-15		PLUMBING INSTL, RIPPLE ATTENUATOR (VICKERS PUMP)							G	
1	65-58145-7		. TUBE ASSY								1
1	65-58145-5		. TUBE ASSY (OPT)								1
-1A	BACN10JM60		. . NUT (U/O 65-58145-7)								2
-1B	BACS13BD60		. . SLEEVE (U/O 65-58145-7)								2
-1C	BACS13AP6		. . SLEEVE (U/O 65-58145-5)								2
-1D	MS21921-6		. . NUT (U/O 65-58145-5)								2
2	65-58145-3		. TUBE ASSY *[1]								
2	65-58145-11		. TUBE ASSY							DFG	1
2	65-58145-11		. TUBE ASSY *[3]								1
2	65-58145-13		. TUBE ASSY *[2]								1
2	65-58145-13		. TUBE ASSY (REPL 65-58145-3)							CDEFG	1
-2A	BACS13AP4		. . SLEEVE (U/O 65-58145-3)								2
-2B	MS21921-4		. . NUT (U/O 65-58145-3)								2
3	10-60563-10		. HOSE ASSY							A	1
			R11547-1, V50599 B316H0671, V98441 95878, V78570; AE702120-10, V00624								
3A	10-60563-35		. HOSE ASSY							A	1
			R11589-1, V50599 96707-35, V78570 (SB 29-1008)								
4	10-60563-13		. HOSE ASSY								1
			R25696-1, V50599 B414E0727, V98441 95880, V78570; AE702120-13, V00624								
5	10-60563-16		. HOSE ASSY							A	1
			R25698-1, V50599 B318D0747, V98441 95882, V78570; AE702120-16, V00624								
6	10-60563-19		. HOSE ASSY							B-F	1
			R11554-1, V50599 96090, V78570								
6A	10-60563-33		. HOSE ASSY (SB 29-1008)							BDF	1
			R11587-1, V50599 96707-33, V78570								

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY	
			1	2	3	4	5	6	7			
1127-6A	10-60563-34		.	H	O	S	E	A	S	S	CEG	
			.	8	3	4	6	H	0	2		
			.	R	1	5	8	8	-	1		
			.	9	6	7	0	7	-	3		
			.	V	7	8	5	7	0			
7	10-60563-20		.	H	O	S	E	A	S	S	BCD	1
			.	R	1	5	5	-	1			
			.	V	5	0	5	9	;			
7	10-60563-42		.	H	O	S	E	A	S	S	E	1
			.	B	4	6	8	E	0	6		
			.	V	9	8	4	4				
			.	R	2	8	1	9	1			
			.	9	7	6	5	1				
7	BACH6R0684DLW		.	H	O	S	E	A	S	S	F	1
			.	S	B	2	9	-	1	0		
7	BACH6R0665DUW		.	H	O	S	E	A	S	S	G	1
			.	S	B	2	9	-	1	0		
7A	AN743-12		.	B	R	A	C	K	E	T	FG	1
			.	S	B	2	9	-	1	0		
7B	BACC1OHS17		.	C	L	A	M	P	*[4]	FG	1
			.	S	B	2	9	-	1	0		
7B	BACC1OEP17		.	C	L	A	M	P	(S	FG	1
			.	S	B	2	9	-	1	0		
			.	*)	[4]						
7C	BACS12CB3-8		.	S	C	R	E	W	(S	FG	1
			.	S	B	2	9	-	1	0		
7D	AN960-10		.	W	A	S	H	E	R	(FG	2
			.	S	B	2	9	-	1	0		
7E	BACN1OJC3		.	N	U	T	(S	B	2	FG	2
			.	S	B	2	9	-	1	0		
8	10-60563-29		.	H	O	S	E	A	S	S	B-G	1
			.	P	R	E	F					
			.	R	2	6	0	2	-	2		
			.	V	5	0	5	9				
			.	B	3	2	2	D	0	7		
			.	1	4							
			.	9	6	1	9	0	-	2		
			.	V	7	8	5	7				
			.	A	E	7	0	2	1	2		
			.	0	6	2	4					
8	10-60563-21		.	H	O	S	E	A	S	S	B-G	1
			.	(O	P	T)				
			.	A	E	7	0	2	1	2		
			.	0	6	2	4					
			.	R	2	6	0	2	-	1		
			.	V	5	0	5	9				
			.	B	3	2	2	D	0	7		
			.	4	2							
			.	9	6	0	9	2				
			.	V	7	8	5	7				
9	AS138-04-0132		.	H	O	S	E	A	S	S		1
			.	(P	O	S	T	S	B		
			.	3	6	-	1	0	0	7		
			.)	(P	R	E	F			
9	16231		.	H	O	S	E	A	S	S		1
			.	(P	R	E	S	B	3		
			.	6	-	1	0	0	7			
9	10642-3		.	H	O	S	E	A	S	S		1
			.	(O	P	T)				
			.	V	9	9	7	5				
10	BACC1OHS12		.	C	L	A	M	P	*[4]		7
			.	S	B	2	9	-	1	0		
10	BACC1OEP12		.	C	L	A	M	P	*[4]		7
			.	S	B	2	9	-	1	0		
10	BACC1ODDC11		.	C	L	A	M	(O	P		7
			.	S	B	2	9	-	1	0		
			.	*)	[4]						
11	BACC1OHS14		.	C	L	A	M	P	*[4]		8
			.	S	B	2	9	-	1	0		
11	BACC1OEP14		.	C	L	A	M	P	*[4]		8
			.	S	B	2	9	-	1	0		
11	BACC1ODDC13		.	C	L	A	M	(O	P		8
			.	S	B	2	9	-	1	0		
			.	*)	[4]						
12	BACC1OHS21		.	C	L	A	M	P	*[4]		6
			.	S	B	2	9	-	1	0		
12	BACC1OEP21		.	C	L	A	M	P	*[4]		6
			.	S	B	2	9	-	1	0		
12	BACC1ODDC21		.	C	L	A	M	P	*[4]		6
			.	S	B	2	9	-	1	0		
12	BACC1OHS16		.	C	L	A	M	P	*[4]		6
			.	S	B	2	9	-	1	0		
12	BACC1OEP16		.	C	L	A	M	(U	S		6
			.	E	D	O	N	6	5	-	5	
			.	8	1	4	5	-	9)		
			.	*)	[4]						
12	BACC1ODDC16		.	C	L	A	M	(O	P		6
			.	T	O	N	6	5	-	5		
			.	8	1	4	5	-	9)		
			.	*)	[4]						
13	BACC1OHS04		.	C	L	A	M	P	*[4]		7
			.	S	B	2	9	-	1	0		
13	BACC1OEP4		.	C	L	A	M	P	*[4]		7
			.	S	B	2	9	-	1	0		
13	BACC1ODDC4		.	C	L	A	M	P	*[4]		7
			.	S	B	2	9	-	1	0		

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1127-14	MS21908-4		.	E	L	B	O	W			2
15	ER10339		.	U	N	I	O	N	, V79470 OR V92003		1
15	2-01066		.	U	N	I	O	N	(OPT), V11328		1
16	AN901-4C		.	G	A	S	K	E	T		1
17	AN960-716		.	W	A	S	H	E	R		2
18	AN924-4		.	N	U	T					2
19	AN924-6		.	N	U	T					1
20	AN924-8		.	N	U	T					1
21	AN924-16		.	N	U	T					1
22	BACE21AW0606		.	E	L	B	O	W		ABCDFG	1
22A	LER1832-6		.	U	N	I	O	N	V11328 (PREF)	BCDF	1
22A	ER1832-6		.	U	N	I	O	N	V79470 (OPT)	BCDF	1
23	307002-S2-6		.	C	O	U	P	L	I	N	1
24	307002-S2-8		.	C	O	U	P	L	I	N	1
25	307002-S5-6		.	C	O	U	P	L	I	N	1
26	307002-S5-8		.	C	O	U	P	L	I	N	1
27	015067-S2-16		.	C	O	U	P	L	I	N	1
28	015067-S5-16		.	C	O	U	P	L	I	N	1
29	BACS12CB3-7		.	S	C	R	E	W			9
30	NAS43HT3-56		.	S	P	A	C	E	R		1
31	BACS12CB3-(-)		.	S	C	R	E	W			AR
31A	AN960D10		.	W	A	S	H	E	R		7
32	NAS679A3W		.	N	U	T					AR
33	NAS43HT3-20		.	S	P	A	C	E	R		3
34	NAS43HT3-32		.	S	P	A	C	E	R		3
35	65-58145-8		.	T	U	B	E	A	S	S	1
35A	BACN10HX16		.	N	U	T					2
36	MS21902-16J		.	U	N	I	O	N		C	1
37	NAS43HT3-56		.	S	P	A	C	E	R	G	1
38	NAS43HT3-32		.	S	P	A	C	E	R	G	3
39	NAS43HT3-24		.	S	P	A	C	E	R	G	1
40	NAS43HT3-72		.	S	P	A	C	E	R	G	1
41	NAS1801-3-()		.	S	C	R	E	W	(SIZE DETERMINED ON INSTALLATION)	G	AR
42	BACC10HS()		.	C	L	A	M	P	*[4]	G	AR
42	BACC10EP-()		.	C	L	A	M	P	(SIZE DETERMINED ON INSTALLATION) *[4]	G	AR
43	BACC10HS04		.	C	L	A	M	P	*[4]		1
43	BACC10EP4		.	C	L	A	M	P	(POST SB 36-1007) *[4]		1
44	BACC10HS07		.	C	L	A	M	P	*[4]		1
44	BACC10EP7		.	C	L	A	M	P	(POST SB 36-1007) *[4]		1
44	BACC10HS09		.	C	L	A	M	P	*[4]		1
44	BACC10EP9		.	C	L	A	M	P	(POST SB 36-1007) *[4]		1
45	BACS12CB3-10		.	S	C	R	E	W	(POST SB 36-1007)		1
46	AN960PD10		.	W	A	S	H	E	R	(POST SB 36-1007)	1
47	BACN10JC3		.	N	U	T			(POST SB 36-1007)		1

OVERHAUL MANUAL

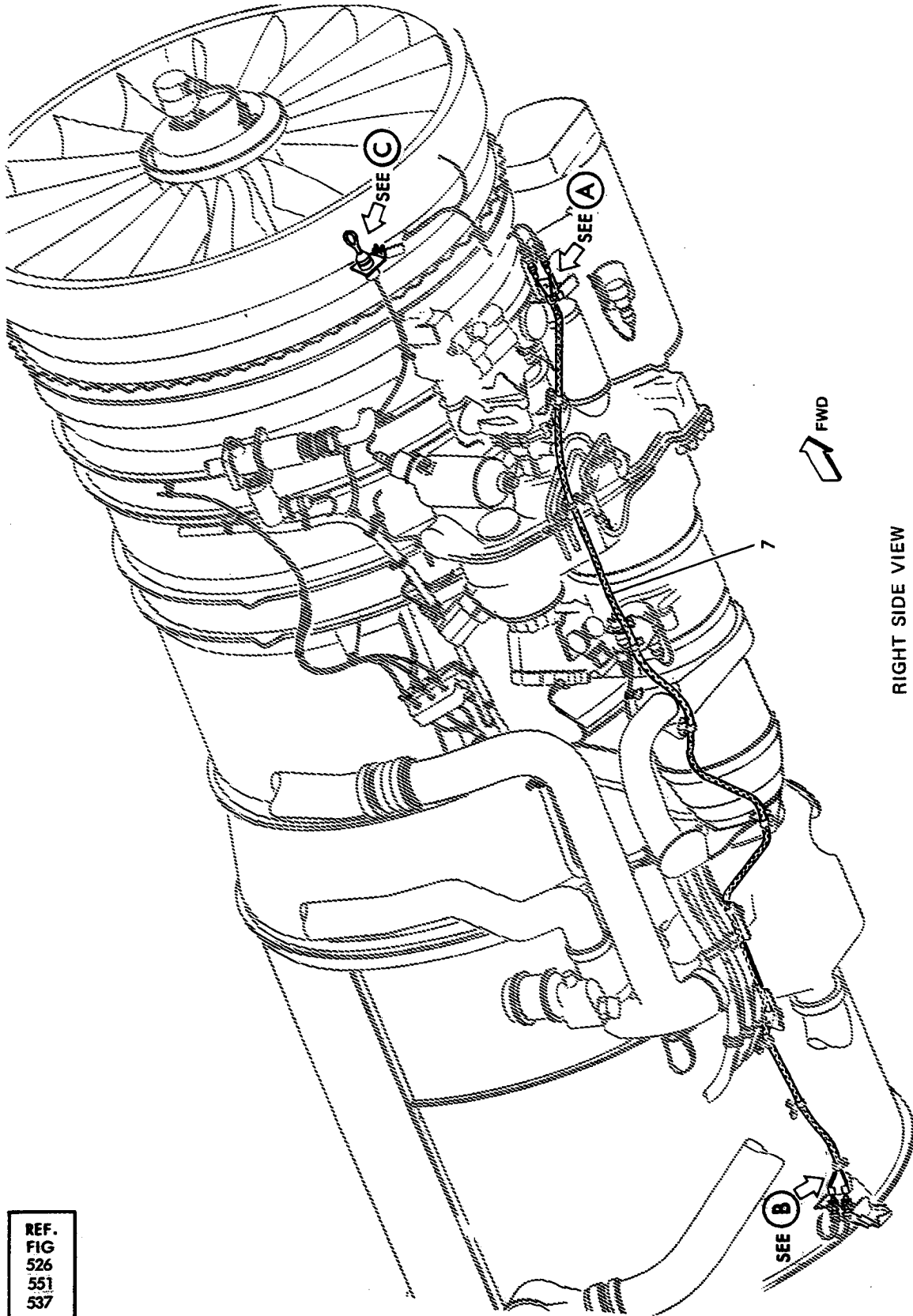
- *[1] PRE SB'S 71-1032, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1079, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151, 71-1179, 71-1189, 71-1198, 71-1278
- *[2] POST SB'S 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1079, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1121, 71-1125, 71-1128, 71-1140, 71-1142, 71-1151, 71-1179, 71-1189, 71-1198, 71-1278
- *[3] POST SB'S 71-1118, 71-1032, 71-1125, 71-1125R1, 71-1125R2, 71-1140, 71-1147, 71-1151, 71-1179
- *[4] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC() CLAMPS

737

BOEING 
COMMERCIAL JET

POWERPLANT BUILDU

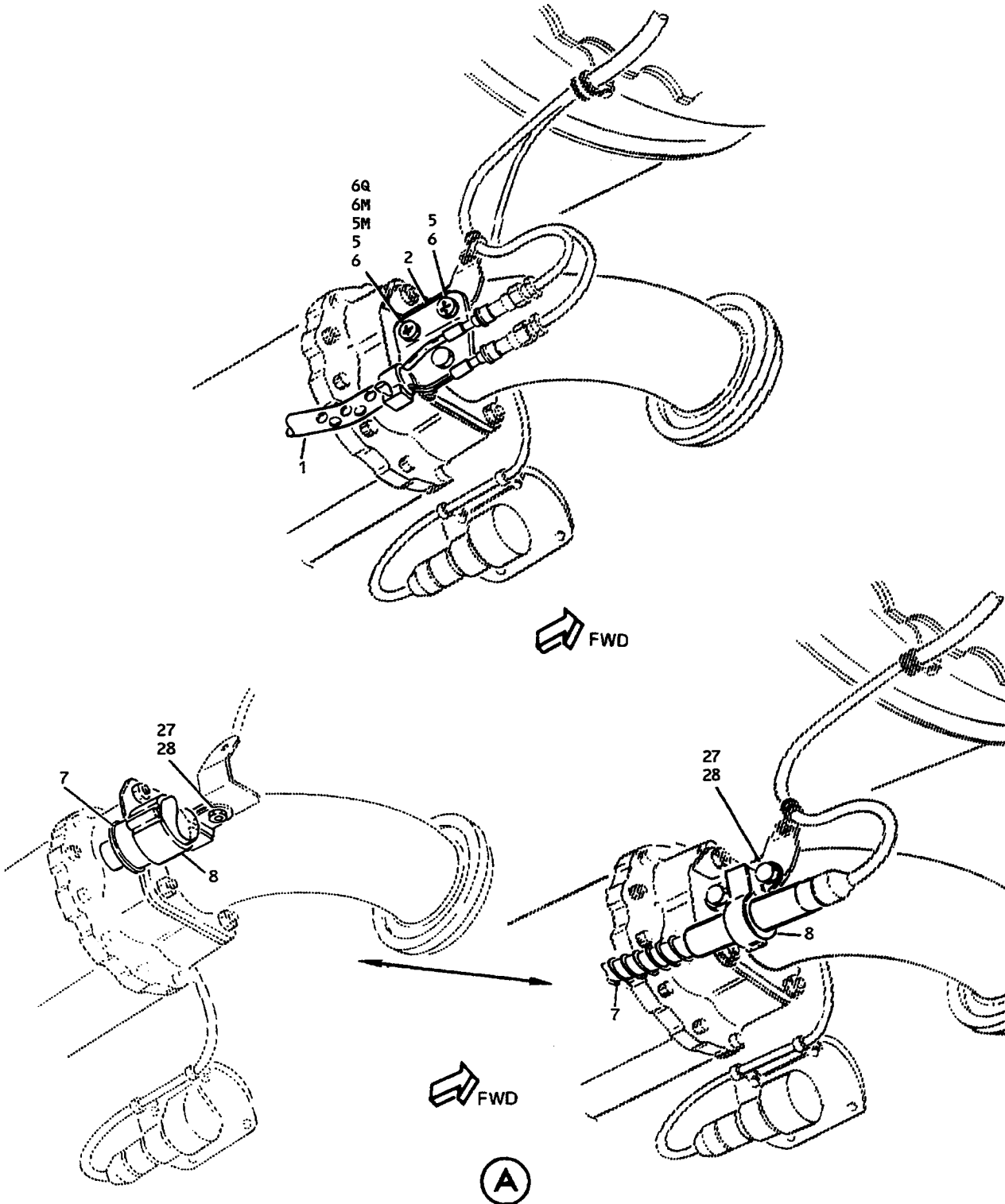
OVERHAUL MANUAL



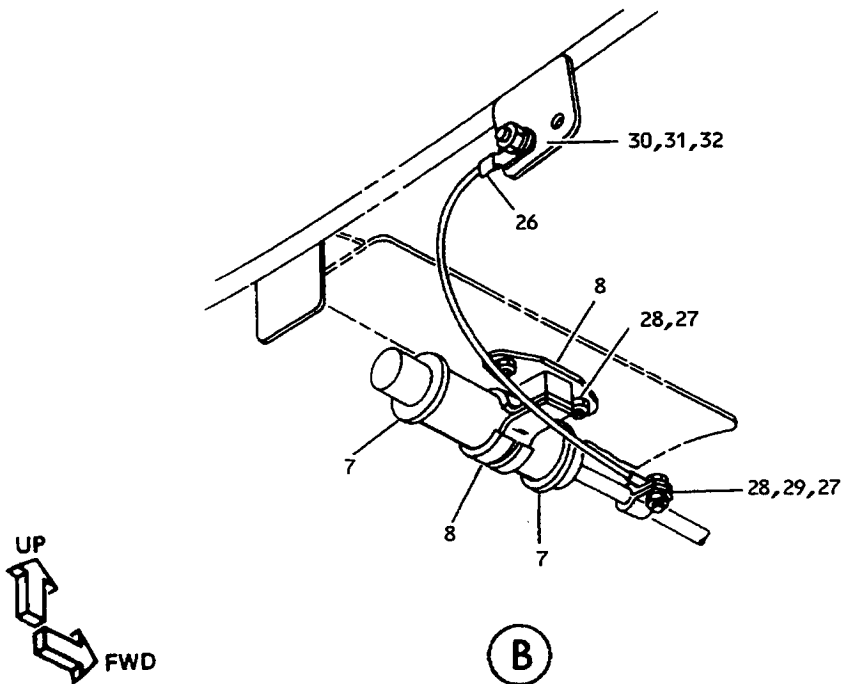
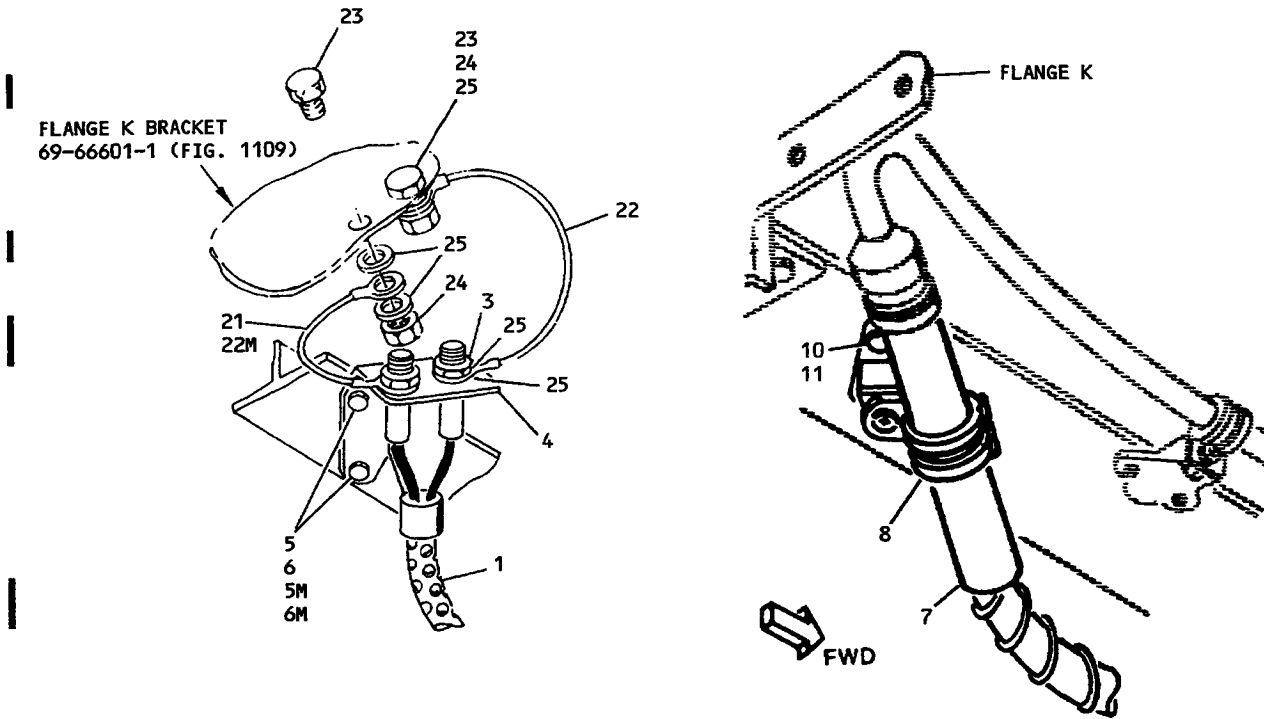
REF.
FIG
526
551
537

Jan 5/80

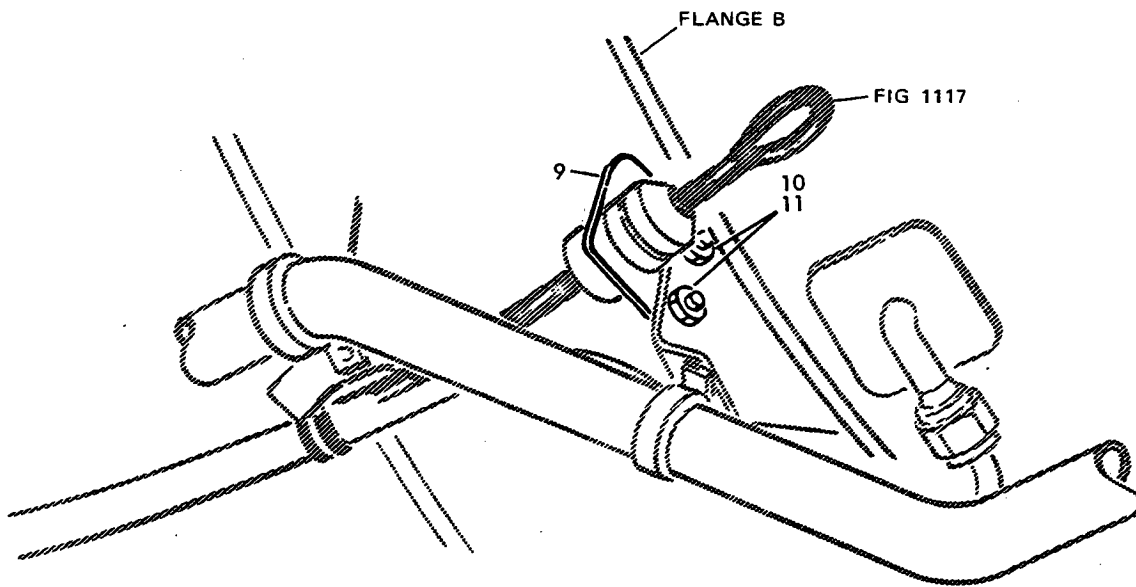
Fire and Overheat Detector Installation
Figure 1128 (Sheet 1)



Fire and Overheat Detector Installation
Figure 1128 (Sheet 2)



Fire and Overheat Detector Installation
Figure 1128 (Sheet 3)



©

 FWD

737


BOEING
COMMERCIAL JET

POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1128-	65-58234-3		DETECTOR INSTL, ENG FIRE & OVERHEAT (KIDDE)							A	RF
	65-58234-5003		DETECTOR INSTL, ENG FIRE & OVERHEAT (KIDDE) (PARTS COLLECTOR, U/O 65-58234-3)							A	RF
	65-58234-9		DETECTOR INSTL, ENG FIRE & OVERHEAT (LINDBERG)							B	RF
	65-58234-5006		DETECTOR INSTL, ENG FIRE & OVERHEAT (LINDBERG) (PARTS COLLECTOR, U/O 65-58234-9)							B	RF
	65-58234-10		DETECTOR INSTL, TAILPIPE EXTENSION (KIDDE)							C	RF
	65-58234-11		DETECTOR INSTL, TAILPIPE EXTENSION (LINDBERG)							D	RF
	65-58234-12		DETECTOR INSTL, TAILPIPE EXTENSION (LINDBERG)							E	RF
	65-58234-15		DETECTOR INSTL, ENG FIRE AND OVERHEAT (SYSTRON-DONNER)							F	RF
	65-58234-16		DETECTOR INSTL, ENG FIRE AND OVERHEAT (SYSTRON-DONNER)							G	RF
	65-58234-17		DETECTOR INSTL, ENG FIRE AND OVERHEAT (SYSTRON-DONNER)							H	RF
	65-58234-18		DETECTOR INSTL, ENG FIRE AND OVERHEAT (SYSTRON-DONNER)							I	RF
	65-58234-19		DETECTOR INSTL, ENG FIRE AND OVERHEAT (KIDDE)							J	RF
	65-58234-5007		DETECTOR INSTL, ENG FIRE & OVERHEAT (KIDDE) (PARTS COLLECTOR, U/O 65-58234-19)							J	RF
	65-58234-20		DETECTOR INSTL, ENG FIRE AND OVERHEAT (KIDDE)							K	RF
1	894093		. ELEMENT, DETECTOR (V33525) (BOEING 10-61096-115) *[3]							A	1
1	899929		. ELEMENT, DETECTOR (V33525) (BOEING 10-61096-315) *[5]							J	1
1	10-61096-615		. ELEMENT, DETECTOR (V33525) *[6]							J	1
1	896798		. ELEMENT, DETECTOR (V33525) (BOEING 10-62061-10)							K	1
2	69-48918-7		. BRACKET *[3]							AJK	1
3	207865		. NUT, CONNECTOR (V33525)							AJK	2
4	69-52283-2		. BRACKET *[3]							A	1
4	69-52283-2		. BRACKET *[3] *[5]							J	1
4	69-52283-4		. BRACKET							K	1
5	BACS12CB3-6		. SCREW							AK	4

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1128-											
5	BACS12CB3-6		.							J	4
5	BACS12CB3-6		.							J	1
5M	NAS1801-3-7		.							J	3
6	NAS679A3W		.							AK	4
6	NAS679A3W		.							J	4
6	NAS679A3W		.							J	1
6M	NAS679A3W		.							J	3
6Q	BACC10GU102		.							J	1
7	1705		.							B	1
7	10-61096-40		.							FH	1
7	10-61096-35		.							GI	1
8	TA5000BH14AT		.							B	2
8	TA5000BH16HB		.							F-I	2
9	69-48918-8		.							B	1
9	69-48918-6		.							B	1
10	BACS12CB3-6		.							B	8
11	BACN10JC3		.							B	8
12	894504		.							C	1
12	1714-600-4		.							DE	1
13	TA12100004		.							C	2
13	TA5000BH8AT		.							CD	2
14	NAS1056E6-031		.							D	4
15	TA5000BH14AT		.							DE	2
15A	65-84112-6		.							E	2
15B	BACS12CB3-8		.							E	2
15C	NAS43HT3-6		.							E	2
15D	AN960-C10		.							E	2
15E	BACN10JC3		.							E	2
16	BACJ40AD67-12		.							C	2
17	AN960C516L		.							C	2
18	207865		.							C	2
19	AN960C616L		.							C	4
20	63-9273-2		.							C	2
21	BACJ40AD57-4		.							AK	1
21	BACJ40AD57-4		.							J	1
22	BACJ40AD57-6		.							AK	1
22	BACJ40AD57-6		.							J	1
22M	BACJ40AD45-6		.							J	1
23	BACB30NE4-1		.							AJK	2
24	BACN10JC4		.							AJK	2
25	AN960C616L		.							AJK	4

737

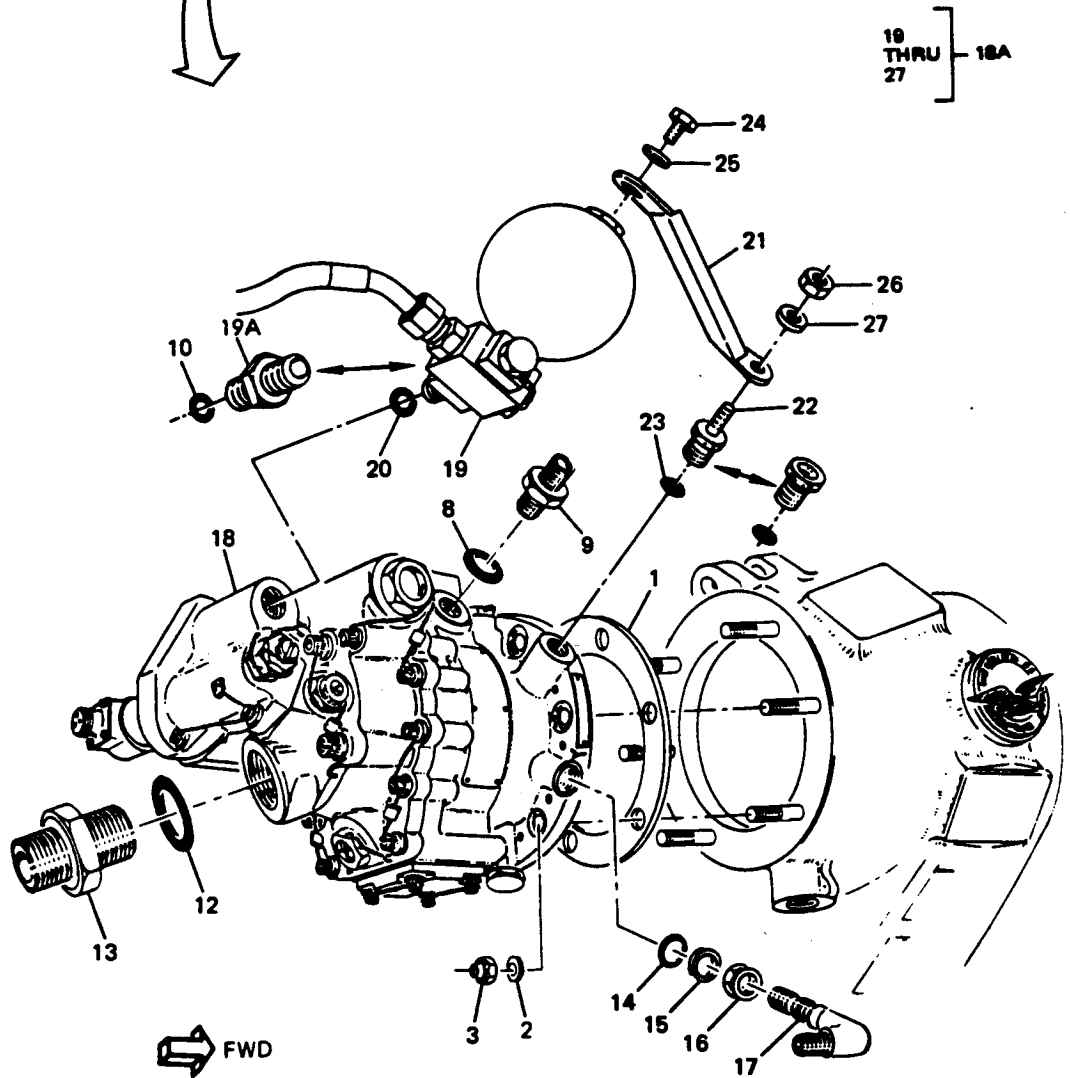
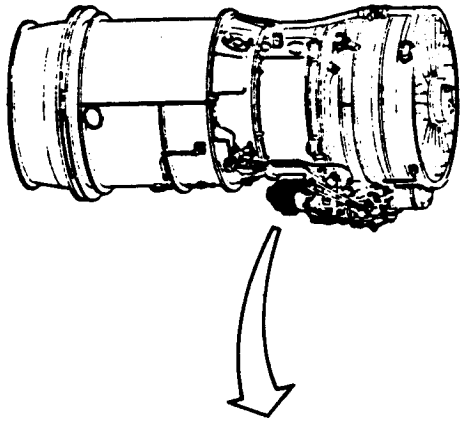
BOEING 
COMMERCIAL JET

POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1128- 25	AN960C416L		.	W	A	S	H	E	R	AJK	4
26	BACJ40AD45- 11		.	J	U	M	P	E	R	F-I	1
27	NAS1801-3-6		.	S	C	R	E	W		F-I	5
28	BACN10JC3		.	N	U	T				F-I	5
29	AN960C10L		.	W	A	S	H	E	R	F-I	1
30	NAS6604-1		.	B	O	L	T			F-I	1
31	AN960C416L		.	W	A	S	H	E	R	F-I	2
32	BACN10JC4		.	N	U	T				F-I	1

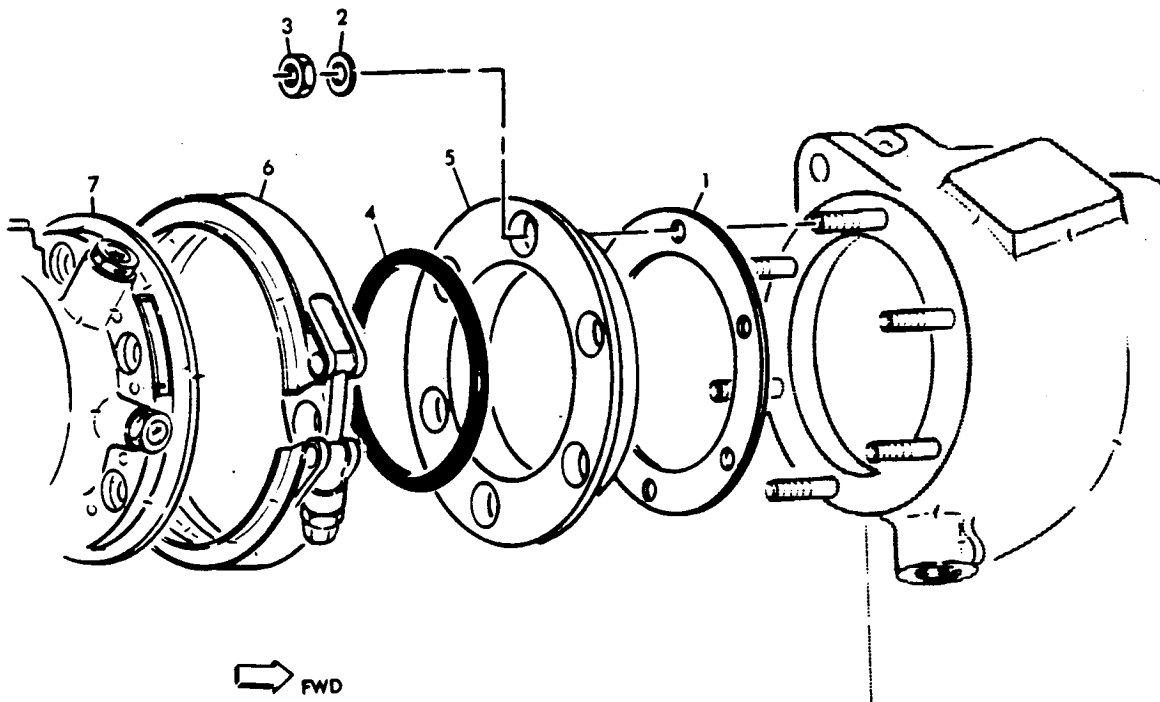
- *[1] POST SB 26-1014
- *[2] PRE SB 26-1017
- *[3] PRE SB 26-1025
- *[4] POST SB 26-1025
- *[5] PRE SB 26-1077
- *[6] POST SB 26-1077



NOTE: FOR ALTERNATE INSTL HAVING INTEGRAL RIPPLE DAMPER,
SEE FIGURE 1130A (SHEET 1)

REF.
FIG
514
530

Engine Driven Hydraulic Pump Installation (Abex/Kellogg)
Figure 1130 (Sheet 1)

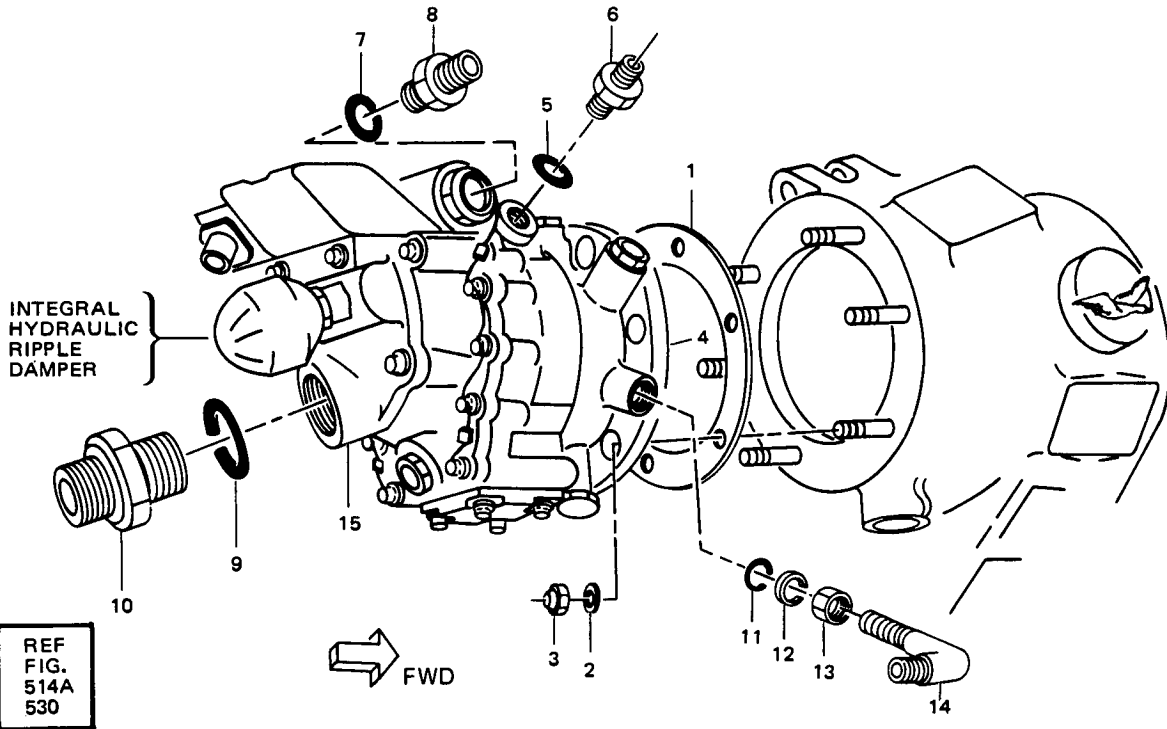


69-15863-10 ASSY

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1130	69-15863-1										
	69-15863-5001										
	69-15863-5002										
	69-15863-5003										
1	AN4047-1										1
-1A	ST1542-01										1
2	AN960D616										6
3	NAS679A6										6
4	5250612										1
4	8023812										1
5	55745										1
6	55759										1
6	22807										1
7	69-15863-8										1
7	69-15863-9										1
7	69-15863-10										1
8	NAS1612-6										1
9	MS21902-6										1
10	NAS1612-10										1
12	NAS1612-20										1
13	AN919-26D										1
14	NAS1612-4										1
15	MS28777-4										1
16	AN6289-4										1
17	BACE21AS0406										1
18	55097										1
18	55098										1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1130 18	65001		. . PUMP, HYDRAULIC, V75250 (BOEING (10-60470-7)(USED ON 69-15863-10)								1
18A	65C13071-1		RIPPLE ATTENUATOR INSTL								
18A	65C13071-2		RIPPLE ATTENUATOR INSTL								
18A	65C13071-3		RIPPLE ATTENUATOR INSTL								
19	9011520100-1		. ATTENUATOR, V83533 *[1]								1
19A	MS21916-10-8		. REDUCER (USED ON 65C13071-3)								1
20	NAS1612-10		. O-RING *[1]								1
21	69-65327-2		. BRACKET (USED ON 10-60470-4 OR -7) *[1]								1
21	69-67501-2		. BRACKET (REPLS 69-65327-2 PER SB 29-1024) (USED ON 65C13071-2)								1
21	69-65327-3		. BRACKET (USED ON 10-60470-5) *[1]								1
21	69-67501-3		. BRACKET (REPLS 69-65327-3 PER SB 29-1024) (USED ON 65C13071-1)								1
21	69-67501-1		. BRACKET (USED ON 65C13071-3)								1
22	66-12113-1		. PLUG *[1]								1
23	NAS1612-4		. O-RING *[1] (USED ON 65C13071-1,-2)								1
24	BACB30NE5H1		. BOLT *[1]								1
24	BACB30NF5H1		. BOLT								1
25	AN960-516		. WASHER *[1]								1
26	BACN10JC4		. NUT *[1]								1
27	AN960-416		. WASHER *[1]								1

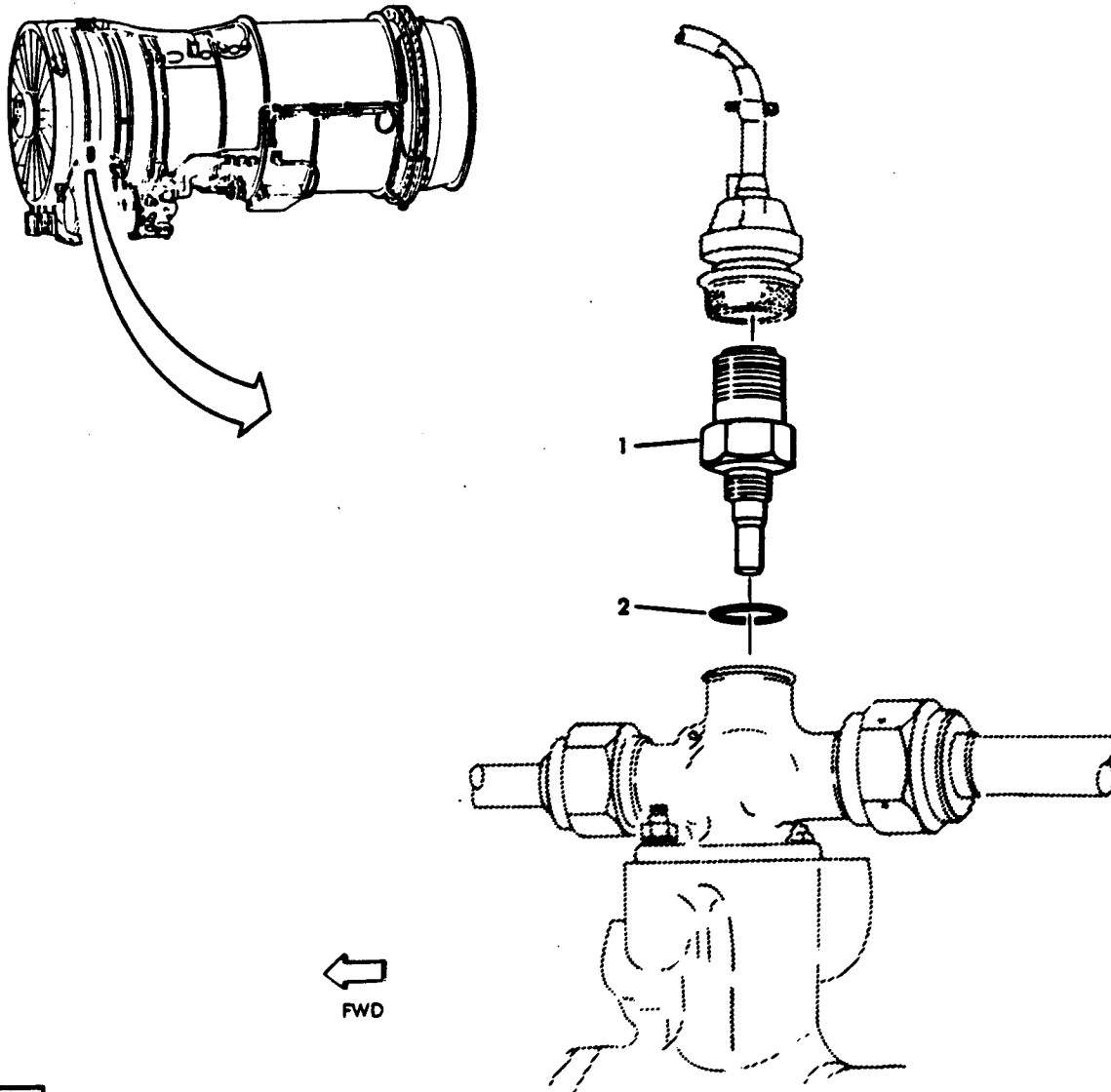
*[1] Added by SB 29-1024



Engine-Driven Hydraulic Pump Installation (Abex)
Figure 1130A (Sheet 1)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1130A	69-15863-5004										
1	AN4047-1										1
-1A	ST1542-01										1
2	AN960D616										6
3	NAS679-A6										6
4	69-15863-21										1
5	NAS1612-6										1
6	MS21902-6										1
7	NAS1612-10										1
8	MS21916-10-8										1
9	NAS1612-20										1
10	AN919-26D										1
11	NAS1612-4										1
12	MS28777-4										1
13	AN6289-4										1
14	BACE21AS0406										1
15	55098-01										1

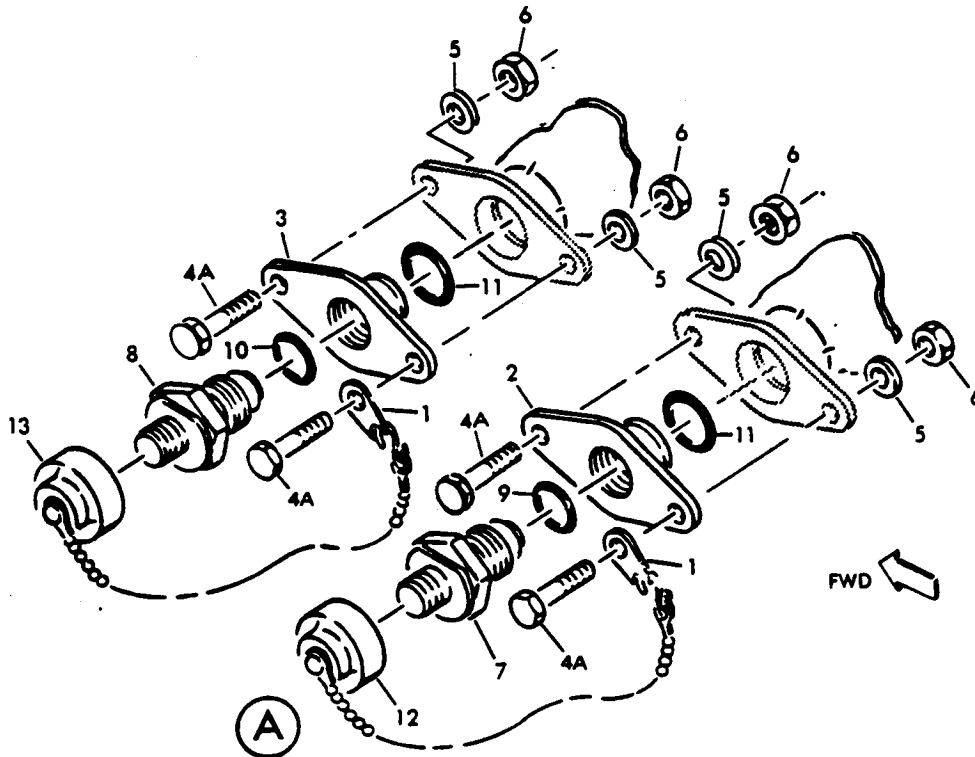
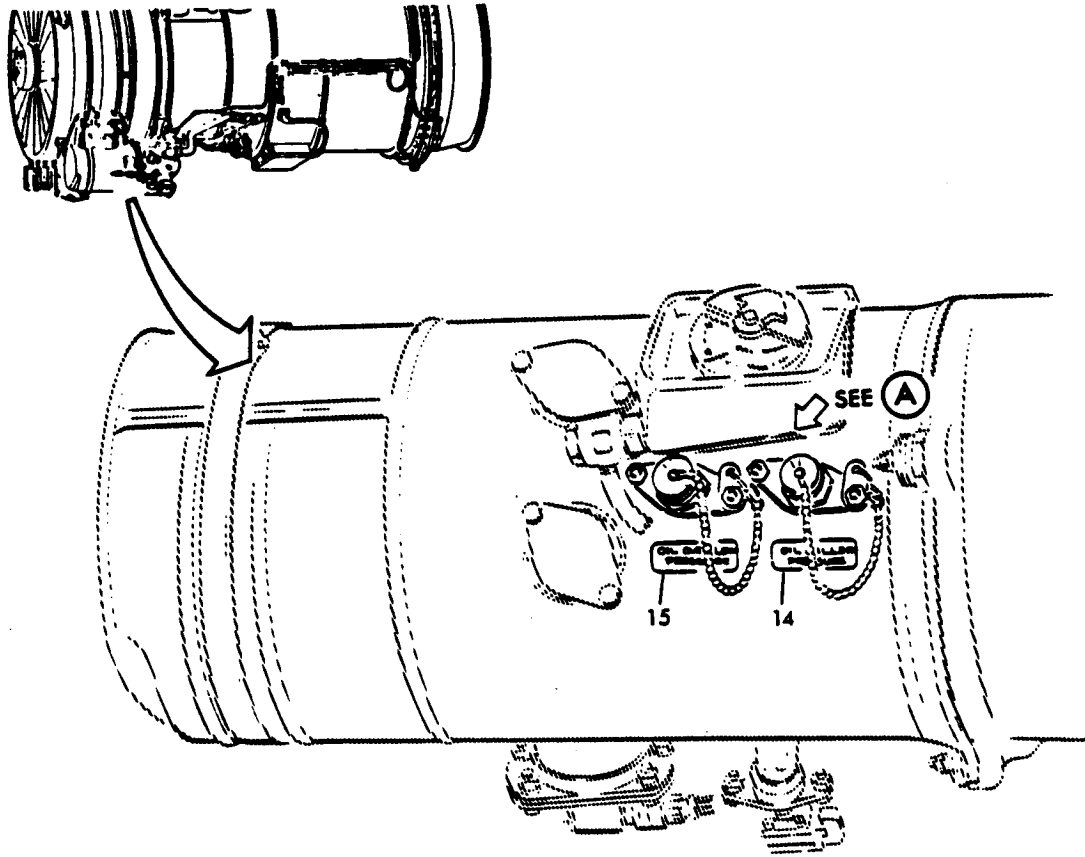
OVERHAUL MANUAL



REF.
 FIG
 546

Engine Oil Temperature Transmitter Installation
 Figure 1131

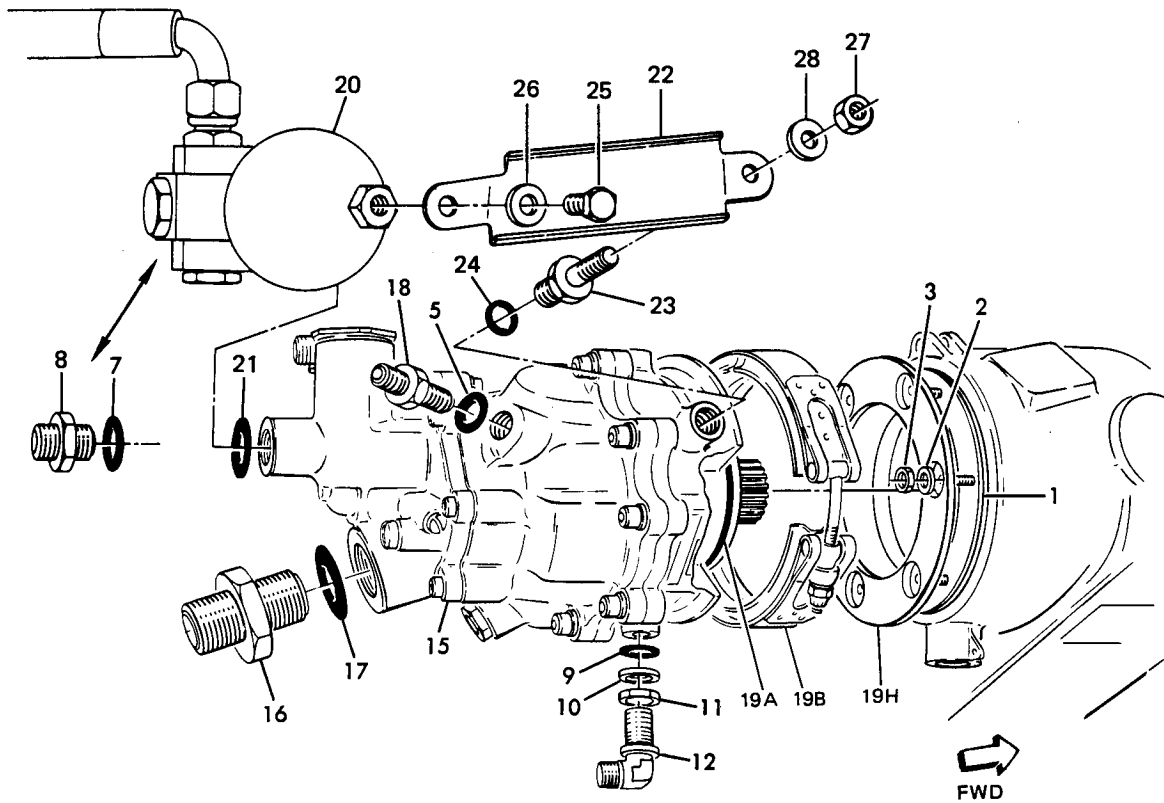
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1131-	69-15935-1		TRANSMITTER INSTL, ENGINE OIL TEMPERATURE								RF
1	56B17D		. OIL TEMPERATURE TRANSMITTER, V35918								1
1	56B17C		. OIL TEMPERATURE TRANSMITTER, V35918 (OPT)								1
2	BACG10U4		GASKET, O-RING								1



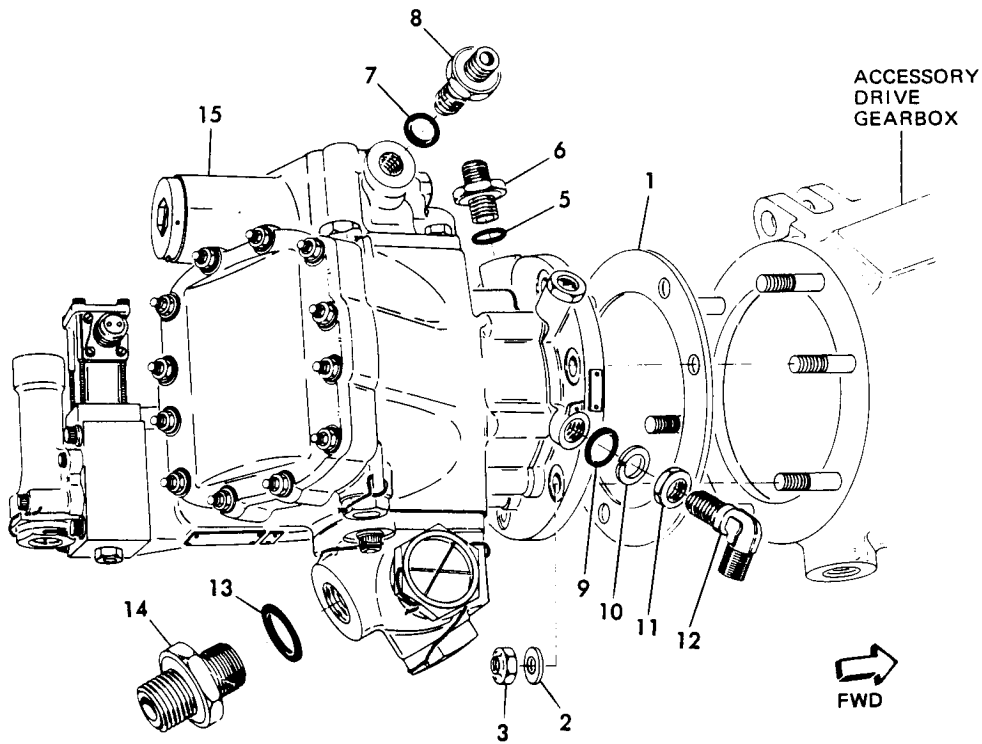
REF.
FIG
540

Engine Oil Tank Pressure Filling Fitting Installation
Figure 1132

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1132-	69-21612-1		FITTING INSTL, OIL TANK PRESSURE FILLING								RF
1	66-16435-1		. CLIP, DUST CAP RETAINING								2
2	69-31661-1		. FITTING, OIL TANK FILLER PRESSURE								1
2	66-16394-1		. FITTING, OIL TANK FILLER PRESSURE (OPT)								1
3	69-31661-2		. FITTING, OIL TANK OVERFLOW PRESSURE								1
3	66-16394-2		. FITTING, OIL TANK OVERFLOW PRESSURE (OPT)								1
4	NAS1104-4		DELETED								
4A	NAS1104-3W		. BOLT								4
5	AN960C416L		. WASHER								4
6	NAS679C4W		. NUT								4
7	OMP2506-2		. COUPLING HALF, BULKHEAD DISCONNECT, V86020								1
8	OMP2505-2		. COUPLING HALF, BULKHEAD DISCONNECT, V86020								1
9	BACP11PB6		. PACKING								1
10	BACP11PB8		. PACKING								1
11	BACP11P21		. PACKING								2
12	OMP2506-4		. CAP ASSY, DUST, V86020								1
13	OMP2505-4		. CAP ASSY, DUST, V86020								1
14	BACM10P6GS		. METAL-CAL, PRESSURE								1
15	BACM10P6GT		. METAL-CAL, OVERFLOW								1



(VICKERS IN-LINE TYPE PUMP)



(VICKERS YOKE-TYPE PUMP)

Hydraulic Pump Installation (Vickers)
Figure 1133

737

POWERPLANT BUILDUP

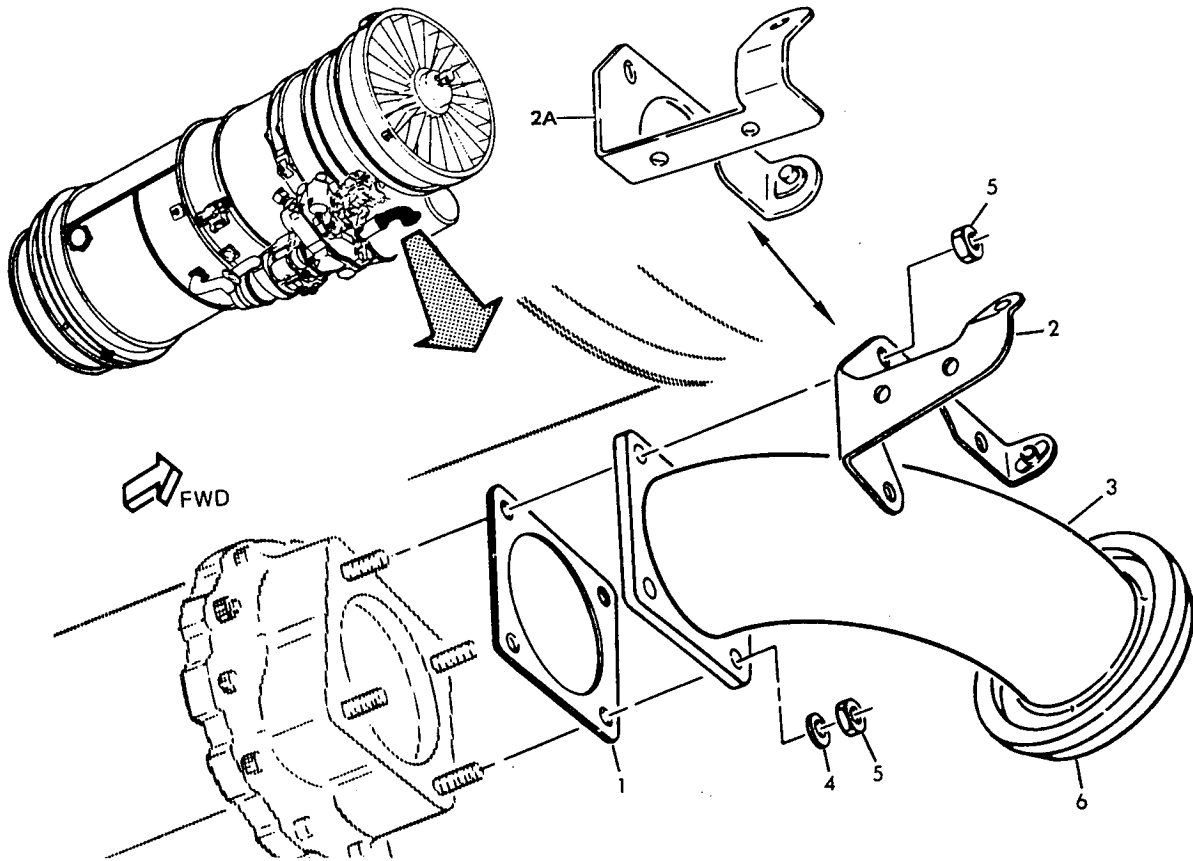


OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1133-	69-28727-1		ENGINE DRIVEN HYDRAULIC PUMP INSTL (VICKERS)								
	69-28727-5001		HYDRAULIC PUMP INSTL (PARTS COLLECTOR, USED ON 69-28727-1)								
	69-28727-5002		HYDRAULIC PUMP INSTL (PARTS COLLECTOR, USED ON 69-28727-1)								
	69-28727-8		HYDRAULIC PUMP INSTL								
1	AN4047-1		. GASKET (SUPSD BY ST1542-01)								1
-1A	ST1542-01		. GASKET (SUPSDS AN4047-1)								1
2	AN960D616		. WASHER								6
3	NAS679A6		. NUT								6
4	69-28727-3		. PUMP ASSY, ENGINE-DRIVEN, HYDRAULIC (USED ON 69-28727-5001)								1
4	69-28727-5		. PUMP ASSY, ENGINE-DRIVEN, HYDRAULIC (USED ON 69-28727-5002)								1
4	69-28727-9		. PUMP ASSY, ENGINE-DRIVEN, HYDRAULIC (USED ON 69-28727-8)								1
5	NAS1612-6		. . PACKING								1
6	MS21924-6		. . UNION (USED ON 69-28727-3 AND -5)								1
7	NAS1612-10		. . PACKING								1
8	MS21916-10-8		. . REDUCER								1
9	NAS1612-4		. . PACKING								1
10	MS28777-4		. . RING								1
11	AN6289-4		. . NUT								1
12	BACE21AS0406		. . ELBOW								1
13	NAS1612-16		. . PACKING (USED ON 69-28727-3 AND -5)								1
14	AN815-16D		. . UNION (USED ON 69-28727-3 AND -5)								1
15	AS66651L8		. . PUMP, HYDRAULIC, V62983 (BOEING 10-60246-8)(USED ON 69-28727-3)								1
15	AS66651L8A		. . PUMP, HYDRAULIC, V62983 (BOEING 10-60246-10)(USED ON 69-28727-5; OPTIONAL ON 69-28727-3)								1
15	PV3-160-4		. . PUMP, HYDRAULIC, V62983 (BOEING 10-61794-1)(USED ON 69-28727-9)								1
16	AN919-26D		. . REDUCER (USED ON 69-28727-9)								1
17	NAS1612-20		. . PACKING (USED ON 69-28727-9)								1
18	MS21902-6		. . UNION (USED ON 69-28727-9)								1
19	MS21902-10		DELETED								
19A	320517		. . . PACKING (PART OF PV3-160-4)								1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1133-19B	374105		.	.	.	CLAMP, MOUNTING (PART OF PV3-160-4)					1
19C	382624		.	.	.	ADAPTER (PART OF PV3-160-4)					1
-19D	387999		.	.	.	ADAPTER AND PINS SUBASSEMBLY (PART OF PV3-160-4)					1
-19F	360527		.	.	.	PIN, ALIGNING					1
-19H	382624		.	.	.	ADAPTER					1
20	9011520100-1		.	.	.	ATTENUATOR, V83533 *[1]					1
21	NAS1612-10		.	.	.	O-RING *[1]					1
22	69-65327-1		.	.	.	BRACKET *[1]					1
22	69-67501-1		.	.	.	BRACKET (REPLACES 69-65327-1 PER SB 29-1024 REV 1)					1
23	66-12113-1		.	.	.	PLUG *[1]					1
24	NAS1612-4		.	.	.	O-RING *[1]					1
25	BACB30NE5H1		.	.	.	BOLT *[1]					1
26	AN960-516		.	.	.	WASHER *[1]					1
27	BACN10JC4		.	.	.	NUT *[1]					1
28	AN960-416		.	.	.	WASHER *[1]					1

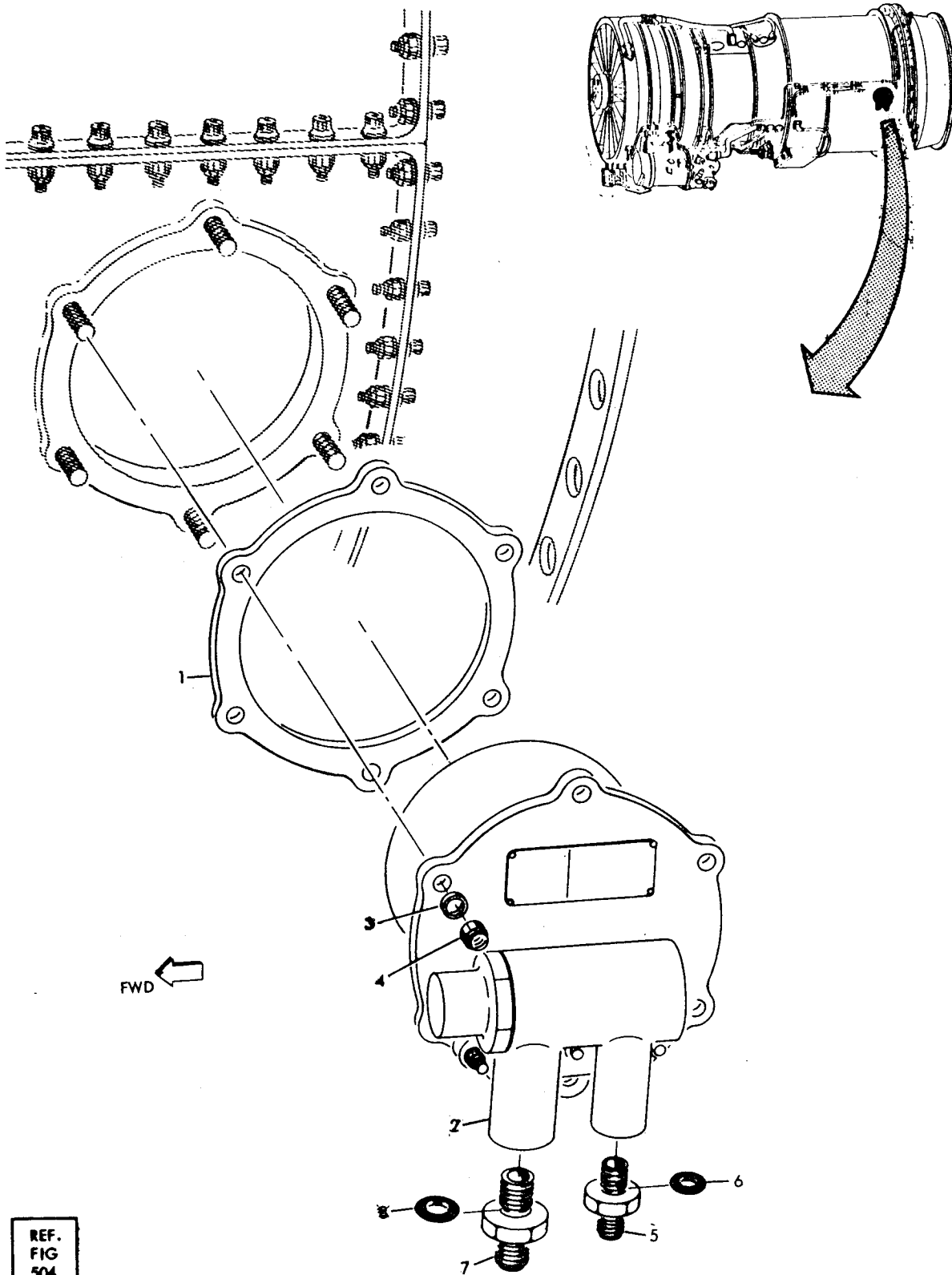
*[1] ADDED BY SB 29-1024 FOR USE ON PUMP ASSEMBLY 69-28727-9



REF.
FIG
537

Fuel Heater Air Exhaust Duct Installation
Figure 1134

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1134-	69-36670-1									A	RF
	69-36670-2									B	RF
1	66-15218-1										1
2	69-52217-4									A	1
2A	69-52217-14									B	1
3	69-36671-1										1
4	AN960C516L									A	1
4	AN960C516L									B	2
5	NAS679C5									A	4
5	BACN10JC5C									B	4
6	69-31608-1										1

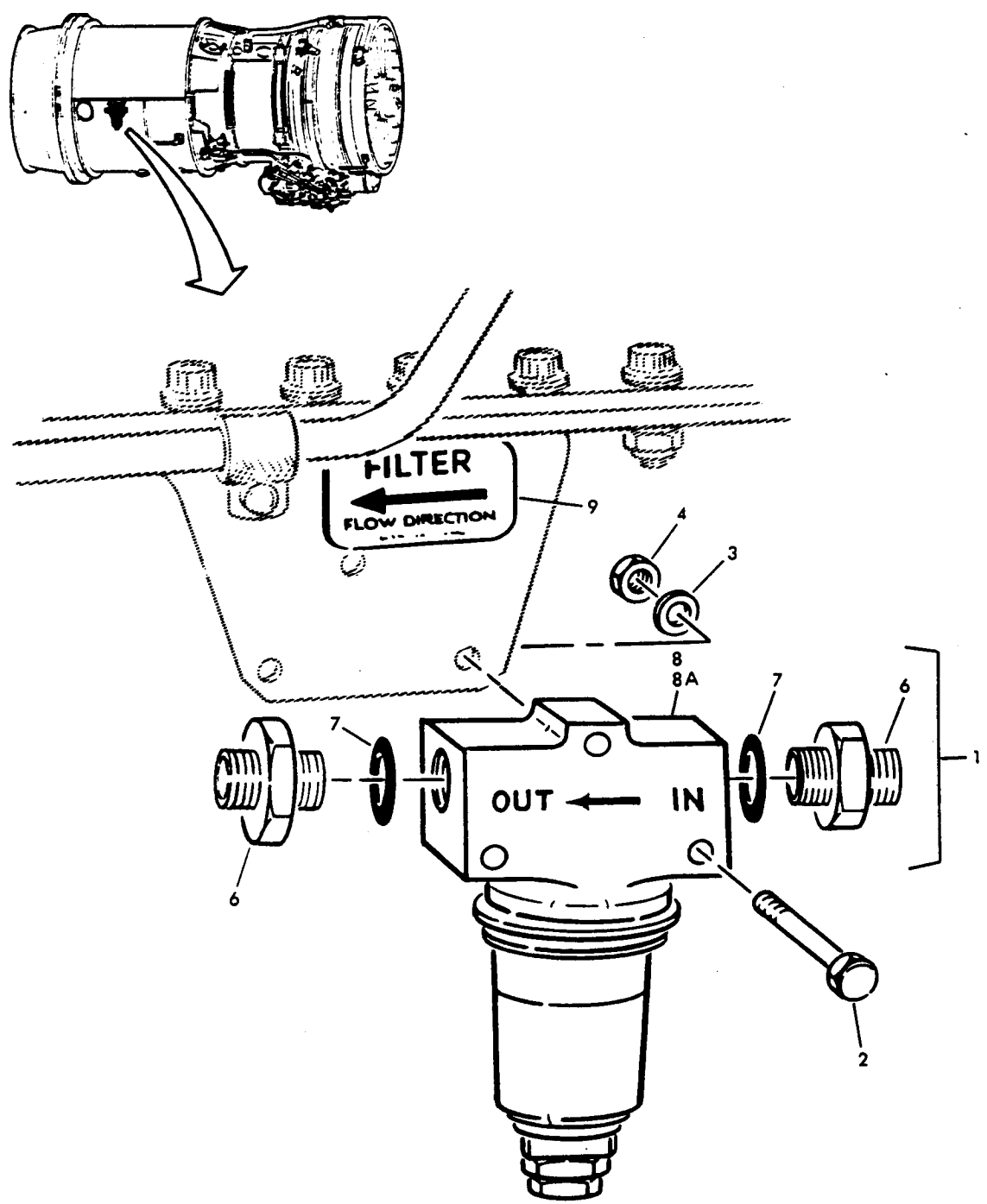


REF.
FIG
504
550

CSD Oil Cooler Installation
Figure 1135

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1135-											
	69-37478-1									A	RF
	69-37478-2									B	RF
1	6A47-003										1
											*[1]
2	6B47-000									A	1
											. OIL COOLER ASSY, CSD, V16630 (BOEING 10-61233-1)*[1]
2	6C47-000									B	1
											. OIL COOLER ASSY, V16630 (BOEING 10-61233-9)
3	AN122582										6
											. WASHER *[1]
4	372B73										6
											. NUT *[1]
5	MS21902-8										1
											. UNION
6	BACP11PB8										1
											. O-RING
7	MS21902-10										1
											. UNION
8	BACP11PB10										1
											. O-RING

*[1] FURNISHED WITH ENGINE

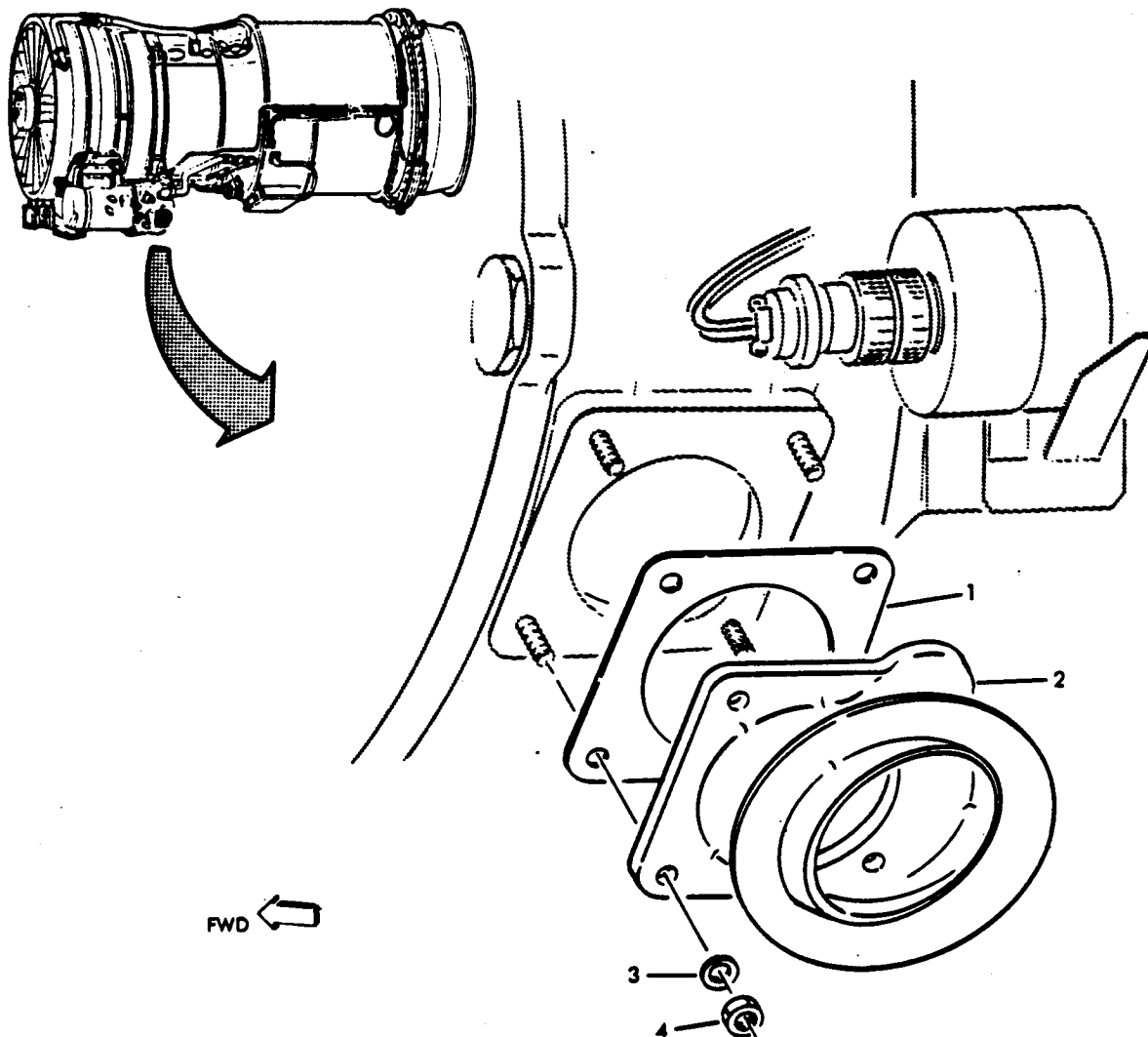


REF.
FIG
516
530

Case Drain Hydraulic Pump Filter Installation
Figure 1136

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1136-	69-45172-1		FILTER INSTL, CASE DRAIN HYDRAULIC PUMP							A	RF
	69-45172-3		FILTER INSTL, CASE DRAIN HYDRAULIC PUMP							B	RF
1	69-45172-2		. FILTER ASSY							A	1
1A	69-45172-4		. FILTER ASSY							B	1
2	NAS1104-30		. BOLT								3
3	AN960D416		. WASHER								3
4	BACN10JC4		. NUT								3
4	NAS679A4W		DELETED								
5	DELETED										
6	MS21902-6		. . UNION								2
7	NAS1612-6		. . PACKING								2
8	053034		. . FILTER (V90005)(BOEING 10-60555-3)							A	1
8A	7585147		. . FILTER (V05228)(BOEING 10-60555-11)							B	1
9	BACM10F11-3AH		. METAL-CAL								1

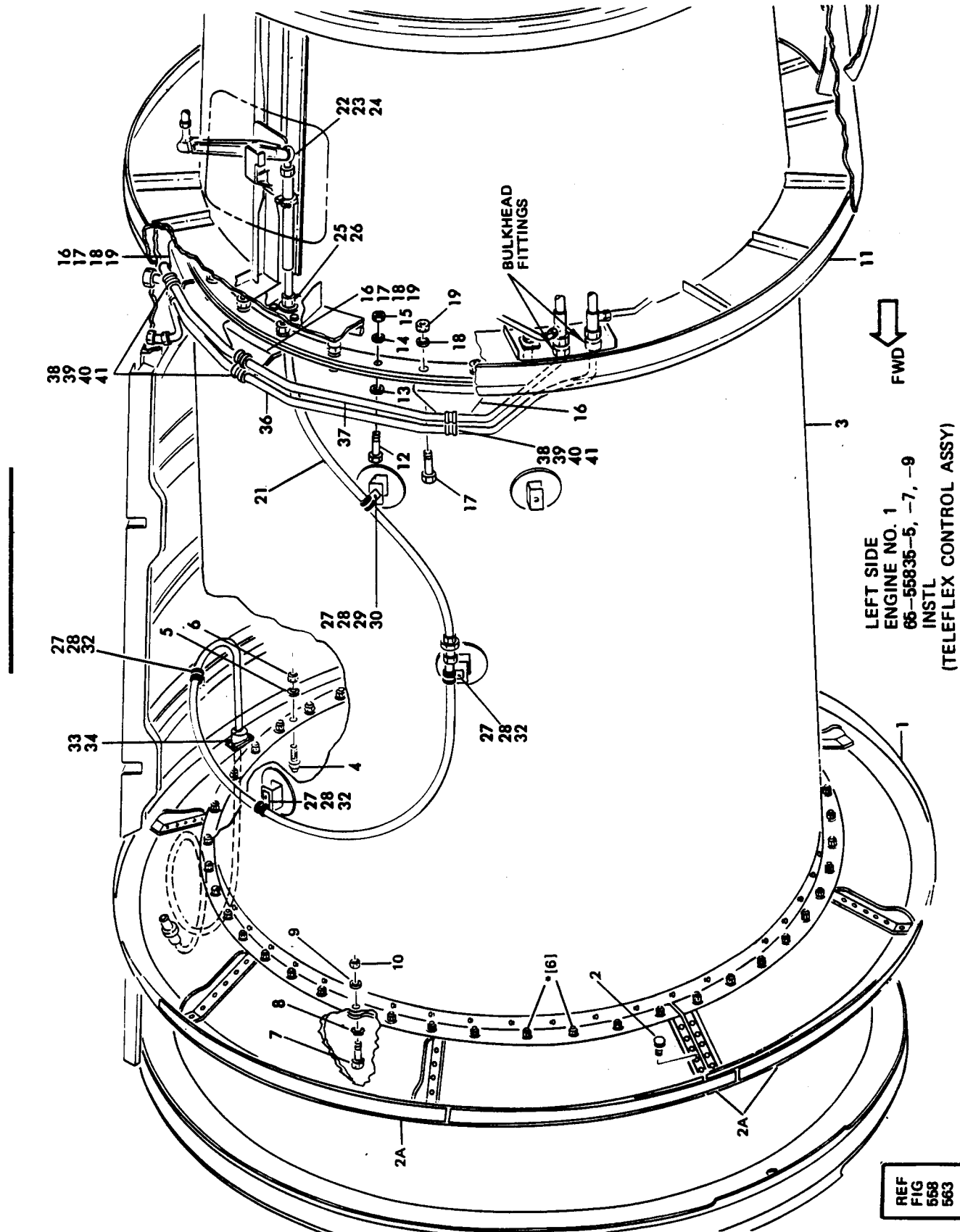
OVERHAUL MANUAL



REF.
 FIG
 541

Engine Gearbox Overboard Breather Fitting Installation
 Figure 1137

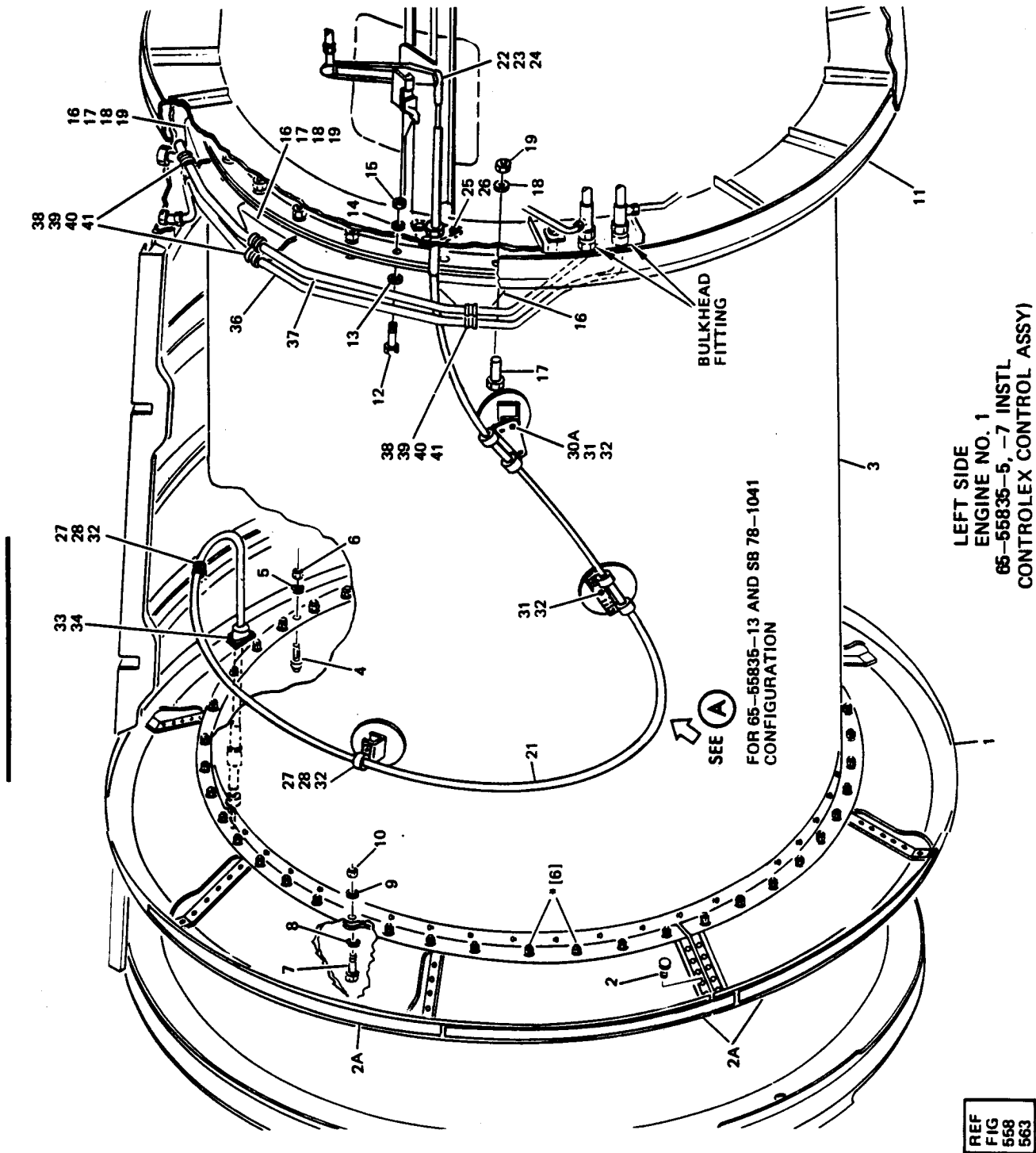
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY. PER ASSY
			1	2	3	4	5	6	7		
1137-	69-48936-1		FITTING INSTL, ENGINE GEARBOX OVERBOARD BREATHER								RF
1	66-14330-1		. GASKET								1
2	65-61732-1		. FITTING ASSY								1
2	65-63375-1		. FITTING ASSY (OPT)								1
3	AN960D416		. WASHER								4
4	NAS679A4W		. NUT								4



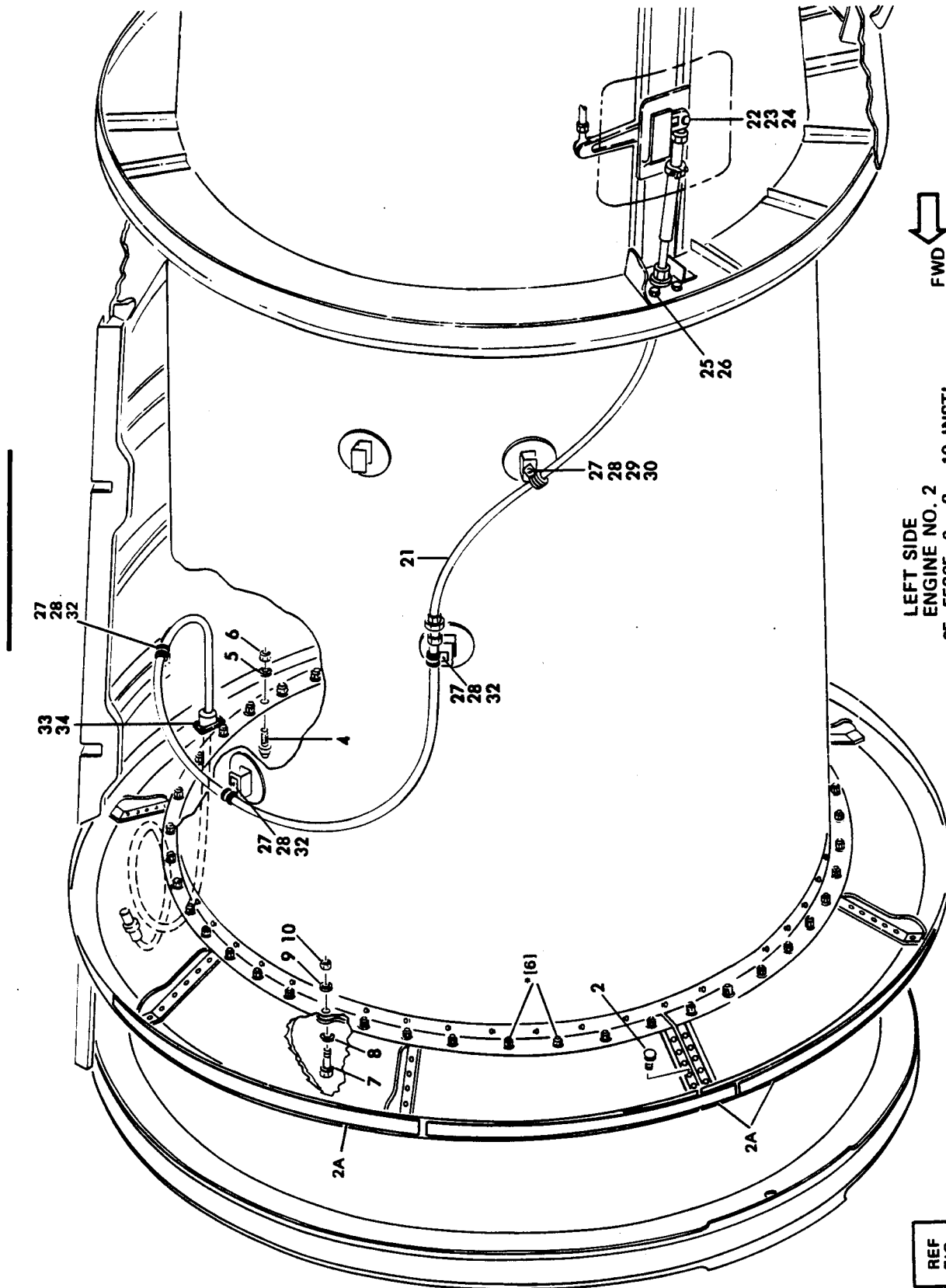
LEFT SIDE
ENGINE NO. 1
66-55835-5, -7, -9
INSTL
(TELEFLEX CONTROL ASSY)

Thrust Reverser, Tailpipe Extension and Shroud Installation
Figure 1138 (Sheet 1)

Dec 5/84



Thrust Reverser, Tailpipe Extension and Shroud Installation
 Figure 1138 (Sheet 1A)

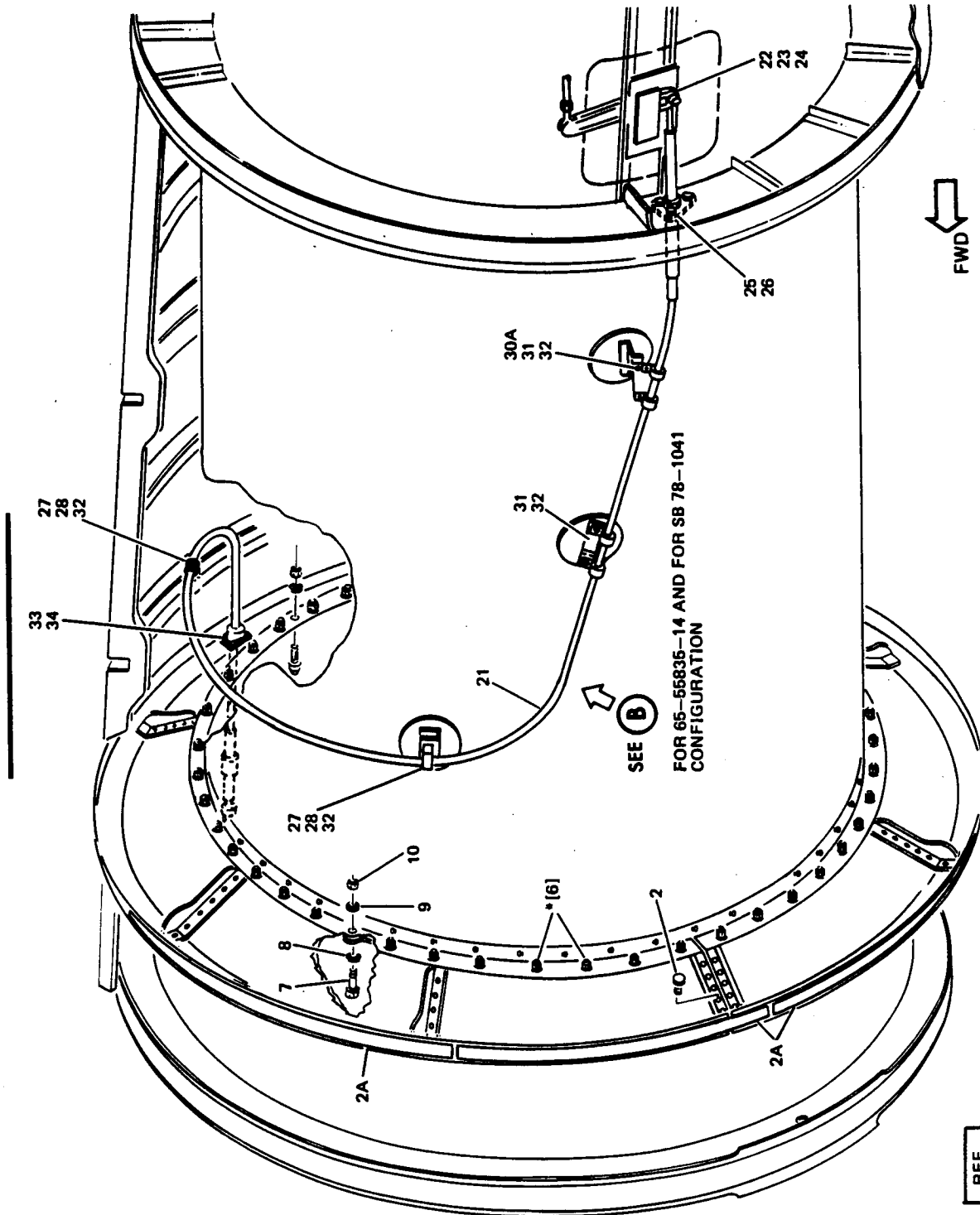


LEFT SIDE
ENGINE NO. 2
65-55835-6, -8, -10 INSTL
(TELEFLEX CONTROL ASSY)

Thrust Reverser, Tailpipe Extension and Shroud Installation
Figure 1138 (Sheet 2)

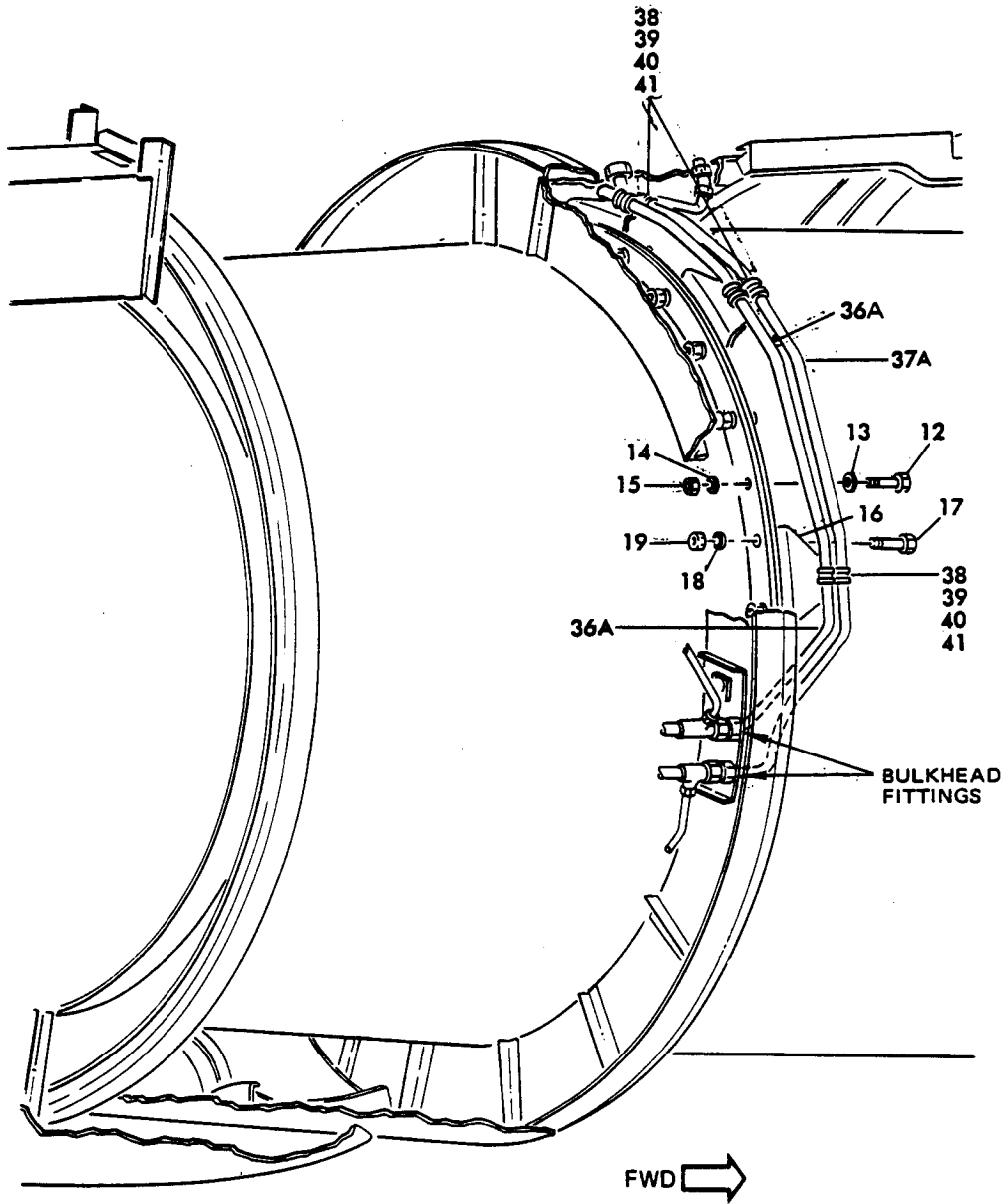
Dec 5/84

71-03-01
Page 1150D1

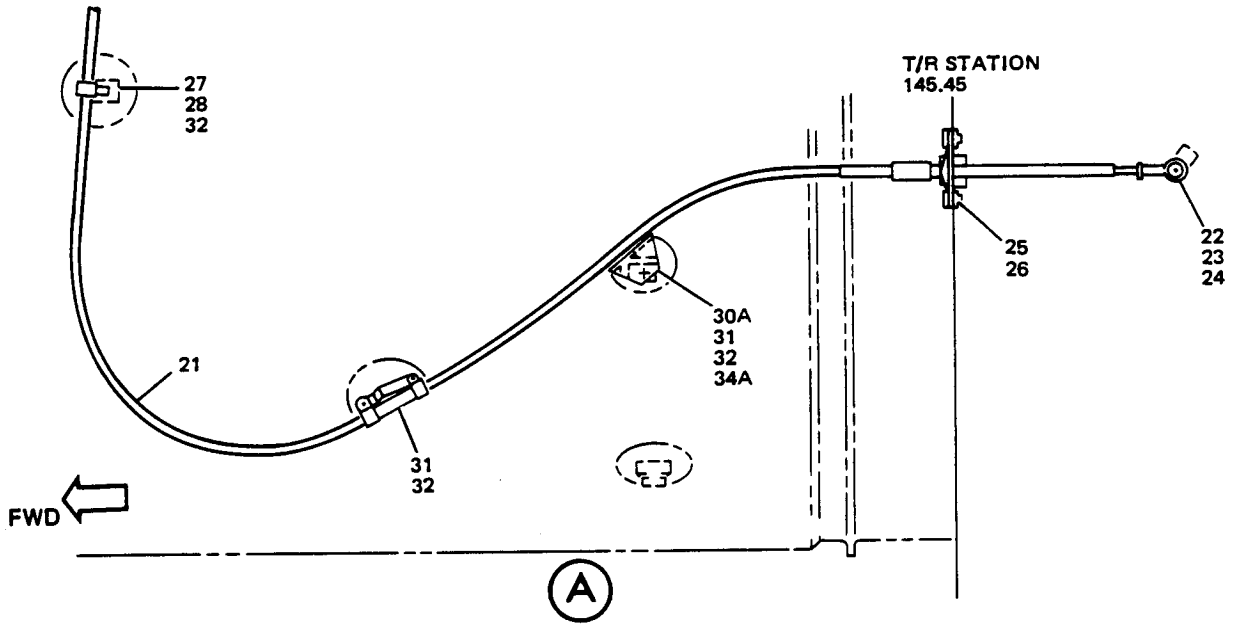


REF	FIG	558	568
-----	-----	-----	-----

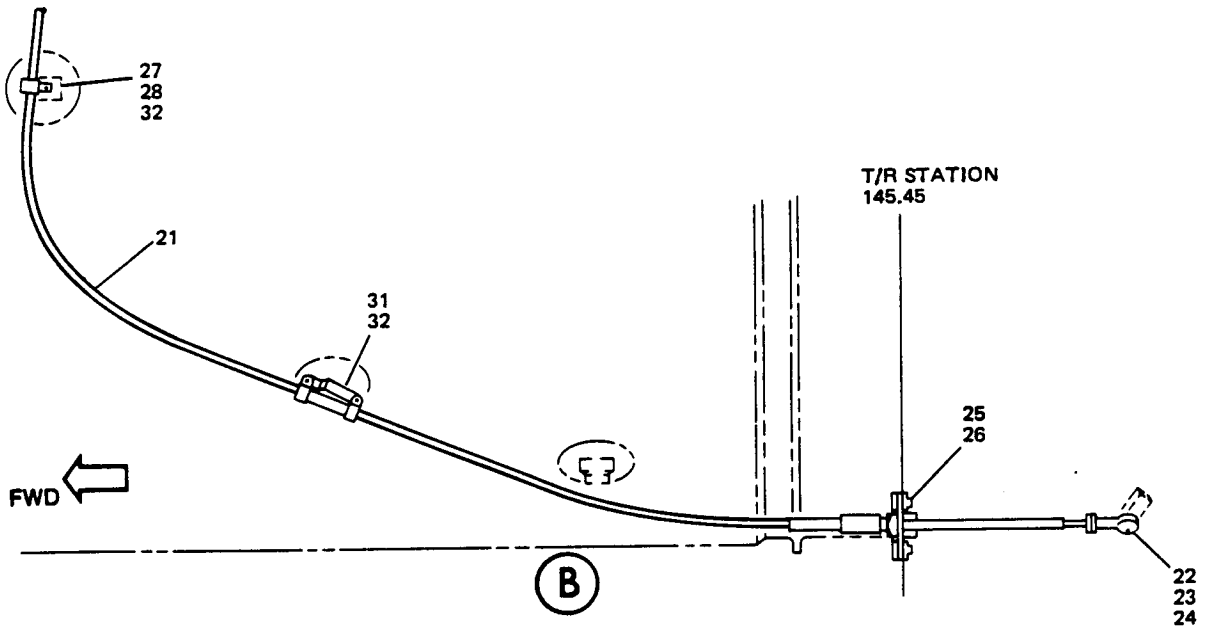
Thrust Reverser, Tailpipe Extension and Shroud Installation
 Figure 1138 (Sheet 2A)



RIGHT SIDE
ENGINE NO. 2



65-55835-13 AND SB 78-1041 CONTROLEX/EANCO PUSH-PULL CABLE CONFIGURATION
ENGINE NO.1



65-55835-14 AND SB 78-1041 CONTROLEX/EANCO PUSH-PULL CABLE CONFIGURATION
ENGINE NO. 2

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1138-	65-70950-1		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1) (SB 78-1005)							A	RF
	65-70950-2		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2) (SB 78-1005)							B	RF
	65-70950-3		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1)							C	RF
	65-70950-4		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2)							D	RF
	65-70950-5		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1)							E	RF
	65-70950-6		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2)							F	RF
	65-70950-7		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1)							G	RF
	65-70950-8		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2)							H	RF
	65-70950-11		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1)							I	RF
	65-70950-12		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2)							J	RF
	65-70950-15		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD ASSY, ENGINE NACELLE							K	RF
	65-70950-16		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD ASSY, ENGINE NACELLE							L	RF
	65-70950-21		THRUST REVERSER, (REPLD BY 65-70950-23)							M	RF
	65-70950-22		THRUST REVERSER, (REPLD BY 65-70950-24)							N	RF
	65-70950-23		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 1) REPLS 65-70950-21)							O	RF
	65-70950-24		THRUST REVERSER, TAILPIPE EXTENSION AND SHROUD INSTL (ENGINE NO. 2) (REPLS 65-70950-22)							P	RF
1	65-70191-1		. SHROUD ASSY, ENGINE NACELLE *[22] *[36]							A-L	1
1	65-70191-34		. SHROUD ASSY, ENGINE NACELLE *[23] *[37]							M-P	1
2	BACB30LK3U2		. . BOLT								21
2A	65-70191-6		. . RUB STRIP								6
3	65-70190-1		. TAILPIPE ASSY, *[4] *[10]							A-D KL	1
3	65-82788-1		. TAILPIPE ASSY, ACOUSTIC, *[4]							E-J OP	1
3	65-82788-1		. TAILPIPE ASSY, ACOUSTIC *[11]*[4]								1

737
POWERPLANT BUILDUP



FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1138-3	65-82788-6		.							E-JOP	1
3	65-82788-16		.							OP	1
4	65-70197-4		.	.							1
5	AN960C10L		.	.							1
6	BACN10GW3A		.	.							1
7	BACB30LE5U6		.							A-L	42
7	BACB30LE5U9		.							M-P	42
8	BACW10BP5ACU		.								42
9	AN960C516		.								42
10	BACN10HR5C		.								42
11	173-1010-1										
11	173-1010-501		.							AE	1
11	173-1011-1		.	.						ABEF	1
11	173-1010-509		.							CG	
11	173-1010-509		.							A	1
11	173-1011-543		.	.						CDGH	1
11	173-1010-2										
11	173-1010-503		.							BF	1
11	65C15618-2		.							AFEF	1
11	173-1010-511		.							DH	1
11	173-1010-511		.							B	1
11	173-1010-517		.							GIK	1
11	173-1010-519		.							HJL	1
11	173-1010-527		.								1
11	173-1010-528		.							N	1
11	173-1010-523		.							O	1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1138-11	173-1010-523		. THRUST REVERSER V51563 (BOEING 10-61813-58, -59)*[2]*[17]							P	1
11	173-1250-1		. THRUST REVERSER *[25]							O	1
11	173-1250-1		. THRUST REVERSER *[25]							P	1
12	BACB30LE5U6		. BOLT *[36]								32
12	BACB30LE5U9		. BOLT *[37]								84
13	BACW1OBP5ACU		. WASHER								32
14	AN960C56		. WASHER								32
14	AN960C516L		. WASHER (OPT)								32
15	BACN1OHR5C		. NUT								32
16	69-49854-1		. BRACKET ASSY *[20]*[36]							A-N	3
16	69-49854-3		. BRACKET ASSY *[21]*[37]							OP	3
17	BACB30LE5U6		. BOLT								6
18	AN960C516L		. WASHER								6
19	BACN1OHR5C		. NUT								6
20	65-55835-5		PUSH-PULL CABLE INSTL (LH) *[12]								RF
20	65-55835-7		PUSH-PULL CABLE INSTL (LH) *[12]								RF
20	65-55835-9		PUSH-PULL CABLE INSTL (LH) *[12]								RF
20	65-55835-6		PUSH-PULL CABLE INSTL (RH) *[12]								RF
20	65-55835-8		PUSH-PULL CABLE INSTL (RH) *[12]								RF
20	65-55835-10		PUSH-PULL CABLE INSTL (RH) *[12]								RF
20	65-55835-13		PUSH-PULL CABLE INSTL (LH) *[12] (REWORKED FROM 65-55835-5)								RF
20	65-55835-14		PUSH-PULL CABLE INSTL (RH) *[12] (REWORKED FROM 65-55835-6)								RF
21	C81662-3		. CONTROL ASSY, AFT, V04638 (BOEING 10-61348-10)(USED ON 65-55835-5 THRU -8)(REPLS 25604 & C81662-2) (POST SB 78-1030) *[40]								1
21	25604		. CONTROL ASSY, AFT, V78710 (BOEING 10-61348-7)(USED ON 65-55835-5 THRU -8)(REPLD BY C81662-3)								1
21	C81662-2		. CONTROL ASSY, AFT, V04638 (BOEING 10-61348-7)(USED ON 65-55835-5 THRU -8)(REPLD BY C81662-3) (PRE SB 78-1030) *[40]								1
21	25604		. CONTROL ASSY, AFT, V78710 (BOEING 10-61348-7)(USED ON 65-55835-9, -10)								1
21	10-61348-14		. CONTROL ASSY, AFT, *[19]*[41]								1
22	BACB30LJ4U10		. BOLT								1
23	AN960416L		. WASHER								1
24	BACN1OJC4C		. NUT								1
25	BACB30LM4HU2		. BOLT								3

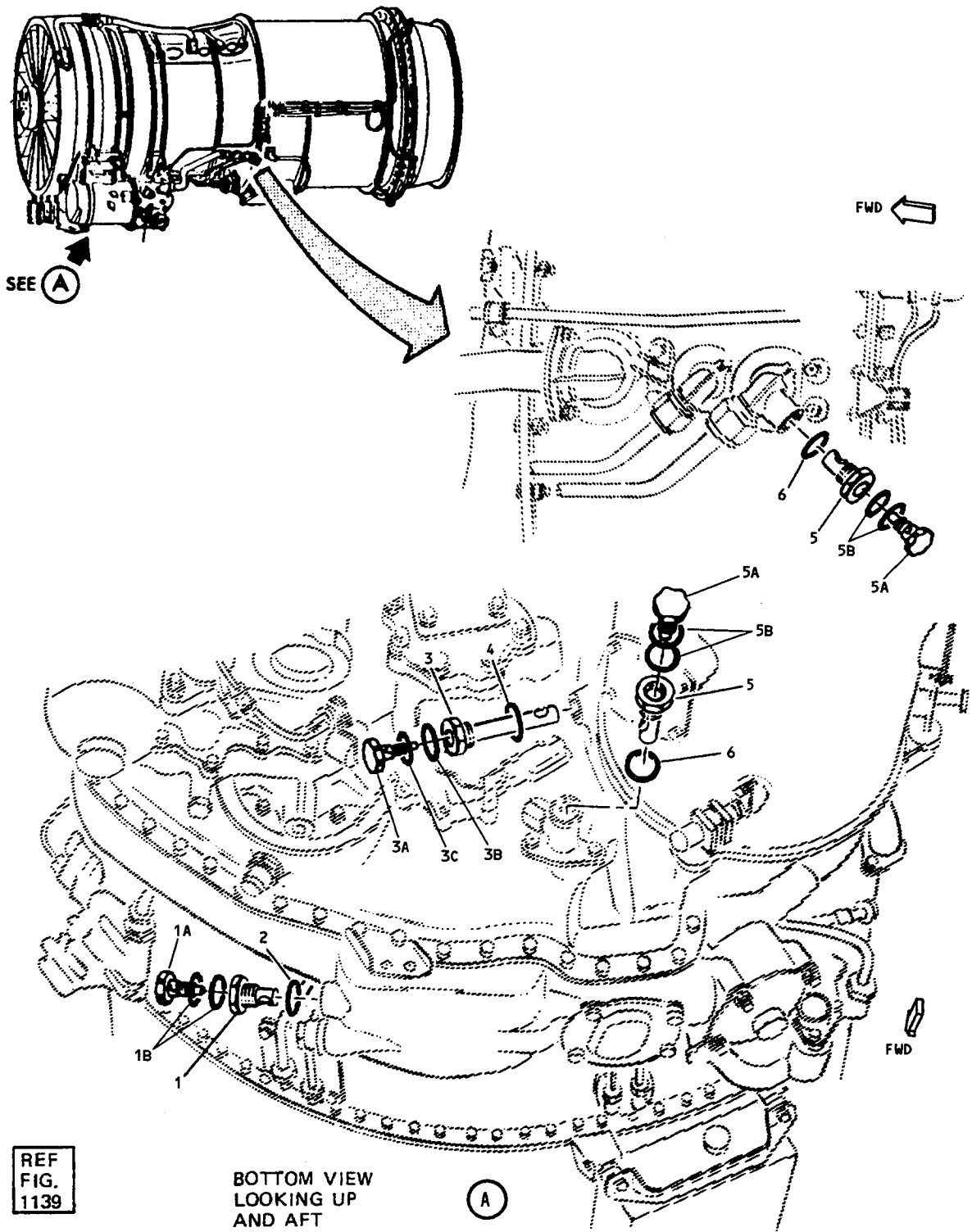
737
POWERPLANT BUILDUP

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1138-											
26	AN960C416L										3
27	BACCL0HS07										4
27	BACCL0EP7										4
27	BACCL0HS07										2
27	BACCL0EP7										2
28	BACB30LM3U2										4
28	BACB30LM3U2										2
29	AN960-10L										1
30	BACN10JC3C										1
30A	BACN10JC3										1
30A	BACN10JC3										1
31	BACB30LM3U3										2
31	BACB30LM3U3										1
32	AN960C10L										3
32	AN960C10L										4
32	AN960C10L										3
33	BACB30LM4U3										2
34	AN960C416L										2
34A	69-77002-1										1
35	65-46858-131										RF
35	65-46858-132										RF
35	65-46858-187										RF
35	65-46858-188										RF
35	65-46858-273										RF
35	65-46858-274										RF
36	65-46858-135										1
36	65-46858-1135										1
36	65C21843-1										1
36A	65-46858-136										1
36A	65-46858-1136										1
36A	65C21843-2										1
37	65-46858-139										1
37	65-46858-1139										1
37	65C21843-3										1

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1138-											
37A	65-46858-140										1
37A	65-46858-1140										1
37A	65C21843-4										1
38	BACCL0HS06										3
38	BACCL0EP6										3
38	BACCL0DDC6										3
38	BACCL0EP26										3
39	BACCL0HS08										3
39	BACCL0EP8										3
39	BACCL0DDC8										3
39	BACCL0EP28										3
40	BACB30LM3U3										3
41	BACN10JC3C										3

- *[1] USED ON JT8D-7, -7A, -9, -9A ENGINES. REF 78-30-12 FOR OVERHAUL.
- *[2] OPTIONAL ON JT8D ENGINES UP TO -15. REF TO 78-30-12 FOR OVERHAUL.
- *[3] USED ON JT8D-17 ENGINES. REF 78-30-12 FOR OVERHAUL.
- *[4] REF 78-38-43 FOR OVERHAUL.
- *[5] USED ON JT8D-15 ENGINE.
- *[6] USE BACB30LE5U8 BOLTS WITH GROUNDING STRAPS USED WITH KIDDE FIRE DETECTOR.
- *[7] SB 71-1055
- *[8] PRE SB 71-1078, 71-1104, 71-1118, 71-1159, 71-1255, 71-1263
- *[9] POST SB 71-1078, 71-1104, 71-1138, 71-1159, 71-1255, 71-1263
- *[10] PRE SB'S 71-1032, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1131, 71-1140, 71-1142, 71-1151, 71-1179, 71-1198, 71-1278
- *[11] POST SB'S 71-1032, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1131, 71-1140, 71-1142, 71-1151, 71-1179, 71-1198, 71-1278
- *[12] LIMITED USAGE
- *[13] PRE SB 71-1086, 71-1088, 71-1138, 71-1258
- *[14] POST SB 71-1086, 71-1088, 71-1258
- *[15] PRE SB 71-1089, 71-1119
- *[16] POST SB 71-1089, 71-1119
- *[17] POST SB 71-1118
- *[18] USED ON TELEFLEX CONTROL ASSY
- *[19] USED ON CONTROLEX CONTROL ASSY
- *[20] PRE SB 71-1139
- *[21] POST SB 71-1139
- *[22] PRE SB 78-1032
- *[23] POST SB 78-1032
- *[24] PRE SB 71-1099
- *[25] POST SB 71-1099
- *[26] PRE SB 71-1119
- *[27] POST SB 71-1119
- *[28] PRE SB 71-1117, 71-1163
- *[29] POST SB 71-1117, 71-1163
- *[30] PRE SB 71-1108, -1150, -1155, -1180, -1337
- *[31] POST SB 71-1108, -1150, -1155, -1180, -1337
- *[32] PRE SB 71-1085, -1089, -1096, -1210
- *[33] POST SB 71-1085, -1089, -1096, -1210
- *[34] PRE SB 71-1154, -1138, -1192
- *[35] POST SB 71-1154, -1138, -1192
- *[36] PRE SB 71-1137, -1162
- *[37] POST SB 71-1137, -1162
- *[38] PRE SB 71-1173, 71-1340
- *[39] POST SB 71-1173, 71-1340
- *[40] PRE SB 78-1041
- *[41] POST SB 78-1041
- *[42] PRE SB 71-1182
- *[43] POST SB 71-1182
- *[44] BACC10HS() REPLACES BACC10EP() AND BACC10EP() REPLACES BACC10DDC() CLAMPS
- *[45] PRE SB 71-1357
- *[46] POST SB 71-1357
- *[47] PRE SB 71-1312, SB 71-1351
- *[48] POST SB 71-1312, SB 71-1351
- *[49] PRE SB 71-1363
- *[50] POST SB 71-1363

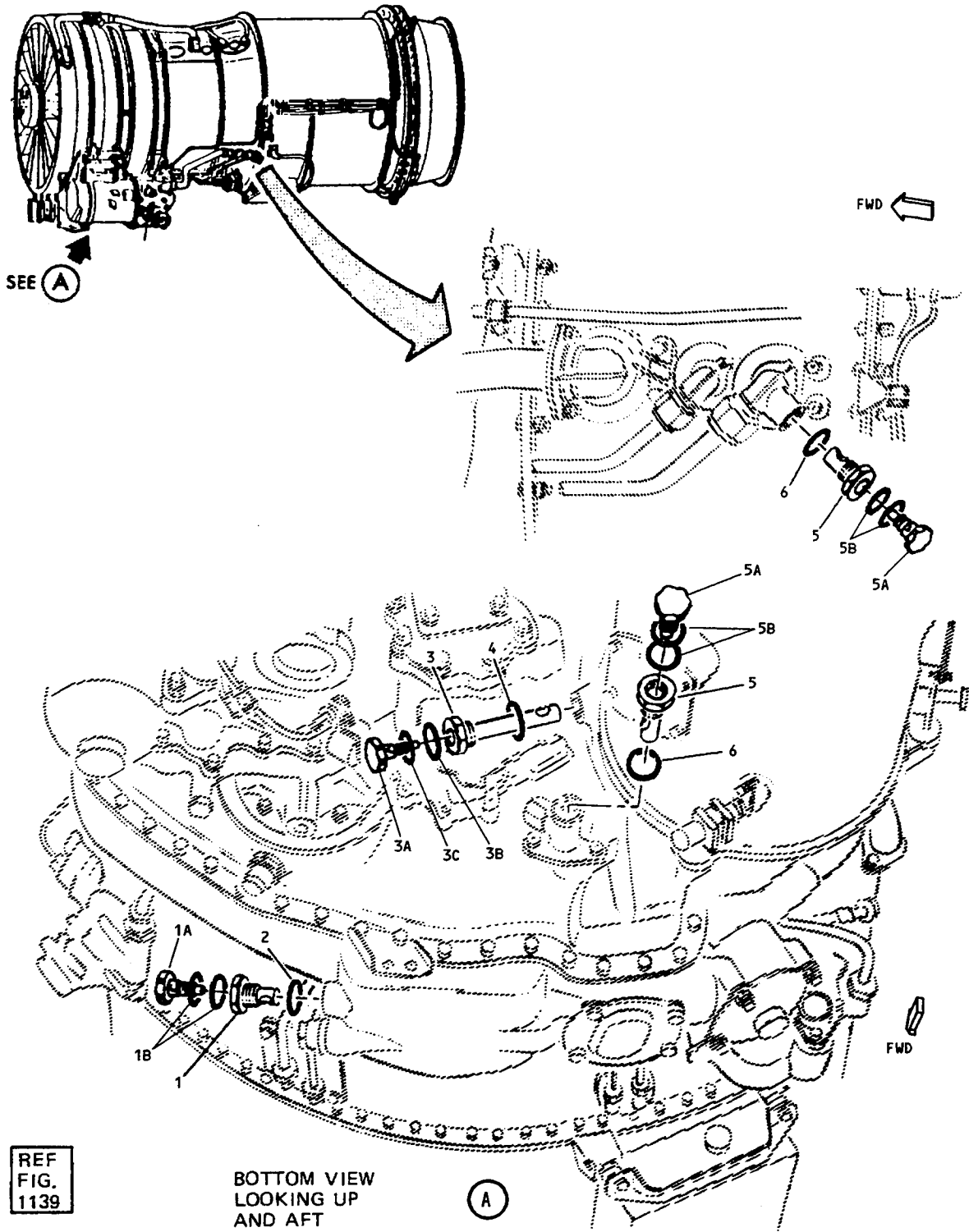


REF
FIG.
1139

BOTTOM VIEW
LOOKING UP
AND AFT

(A)

Chip Detector Installation
Figure 1139



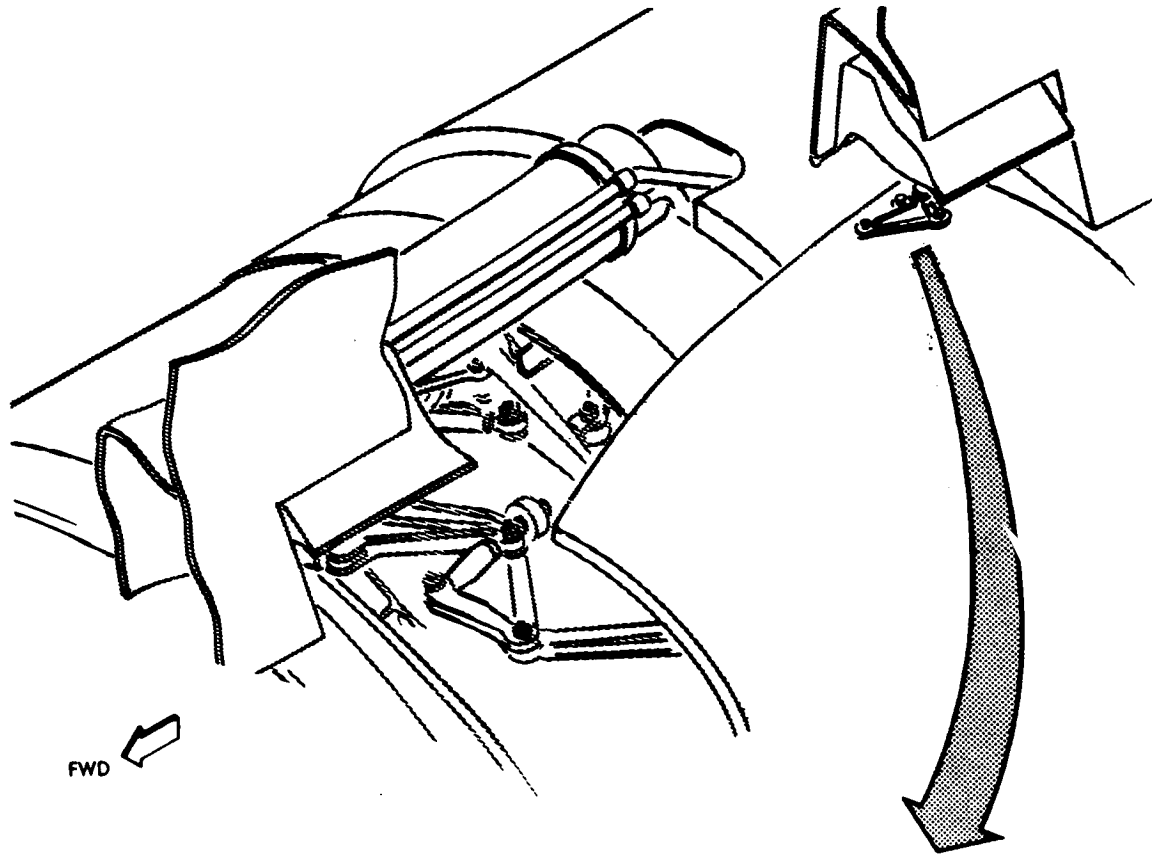
REF
FIG.
1139

BOTTOM VIEW
LOOKING UP
AND AFT

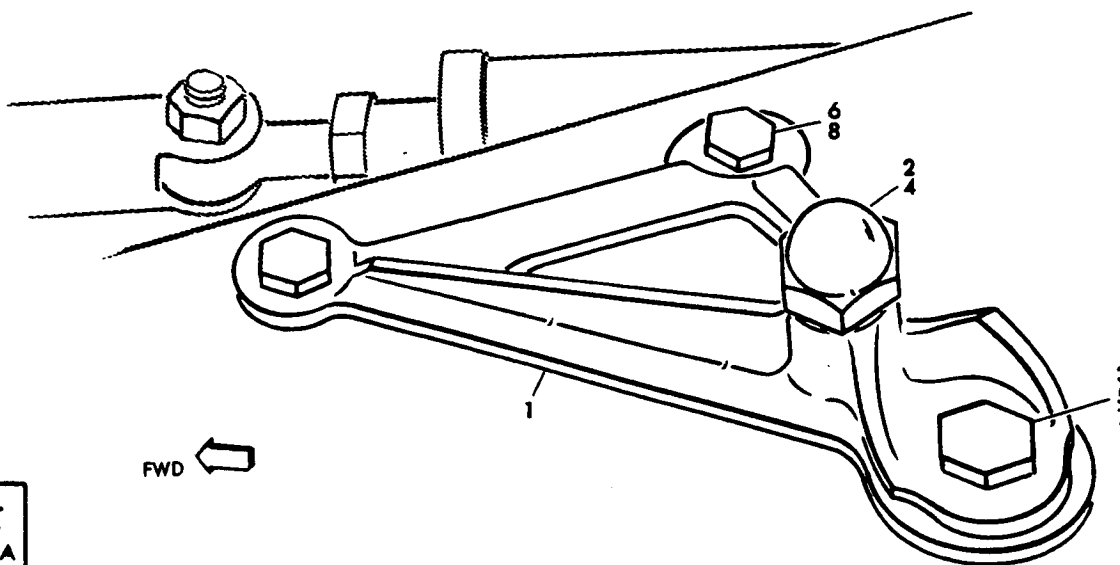
Chip Detector Installation
Figure 1139

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1139	69-56334-1		CHIP DETECTOR INSTL								RF
1	B6003		. TURBOMAG PLUG, V97484								1
1A	B6032		. . MAGNETIC PLUG, V97484								1
1B	M83248/1-012		. . PACKING, O-RING								2
2	BACP11P-B8A		. PACKING, O-RING								1
3	B6002		. TURBOMAG PLUG, V97484.								1
3A	B6035		. . MAGNETIC PLUG, V97484								1
3B	M83248/1-008		. . PACKING, O-RING								1
3C	M83248/1-011		. . PACKING, O-RING								1
4	BACP11P-B6A		. PACKING, O-RING								1
5	B7670		. TURBOMAG PLUG, V97484								2
5A	B6011		. . MAGNETIC PLUG, V97484								2
5B	M83248/1-008		. . PACKING, O-RING								4
6	BACP11P-B5A		. PACKING, O-RING								2



65-57815-1 INSTALLATION AS SHOWN
65-57815-2 INSTALLATION OPPOSITE



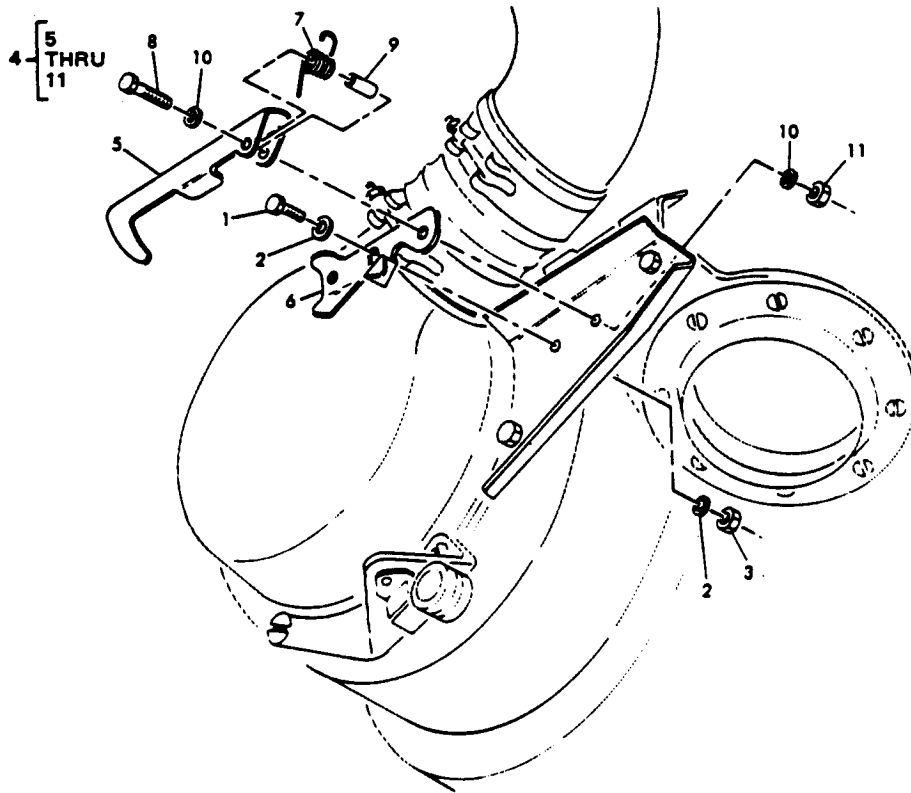
REF.
FIG
566 A
571 A

Thrust Reverser Flipper Door Striker Installation
Figure 1140

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1140-	65-57815-1		STRIKER INSTL, THRUST REVERSER FLIPPER DOOR							A	RF
	65-57815-2		STRIKER INSTL (OPP 65-57815-1)							B	RF
1	65-57814-1		. BRACKET (OPP 65-57814-2)(OPT TO 65-57814-5)							A	1
1	65-57814-5		. BRACKET (OPP 65-57814-6)(OPT TO 65-57814-1)							A	1
1	65-57814-2		. BRACKET (OPP 65-57814-1)(OPT TO 65-57814-6)							B	1
1	65-57814-6		. BRACKET (OPP 65-57814-5)(OPT TO 65-57814-2)							B	1
2	69-49832-1		. STRIKER								1
3	AN4C14A		. BOLT								2
4	AN960C516L		. WASHER								4
5	NAS679C4W		. NUT, SELF-LOCKING *[00]								1
6	AN3C4A		. BOLT (PREF)								2
6	BACB30FM6A3U		. BOLT *[00]								2
7	BACW10P186C		. WASHER *[00]								1
8	BACC3026		. COLLAR *[00]								2

*[00] LIMITED USAGE

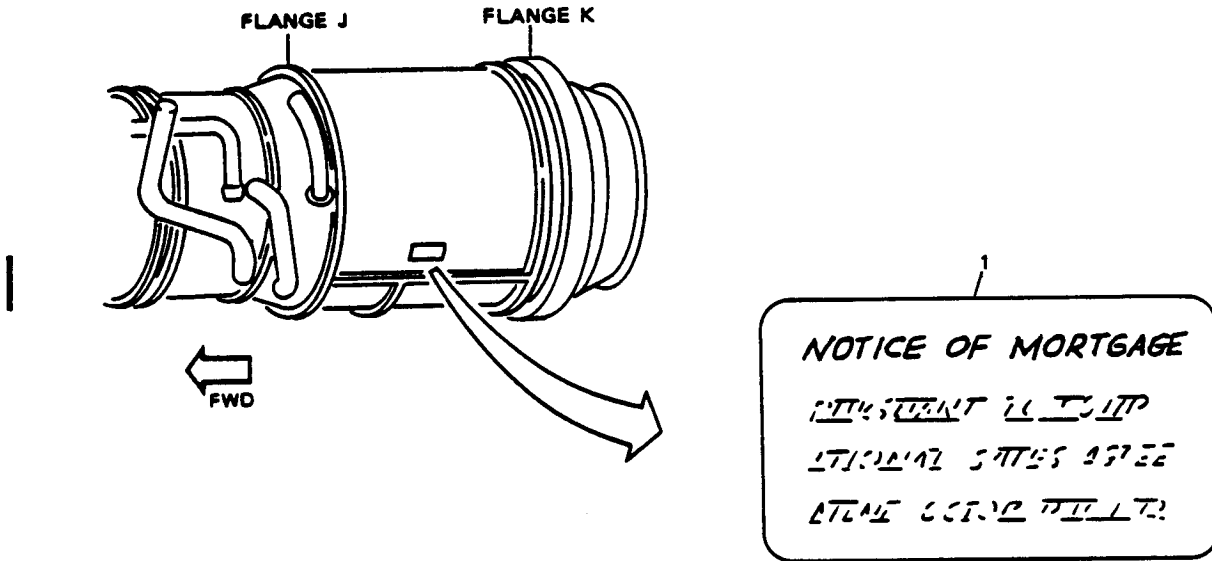
OVERHAUL MANUAL



REF.
FIG
523
1122

Cowl Panel Hook Installation (LH)
Figure 1141

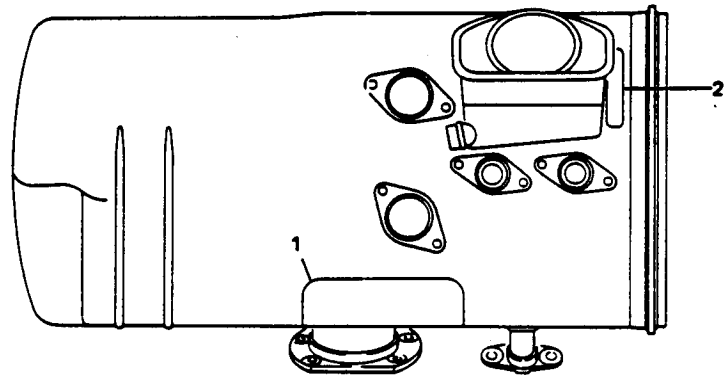
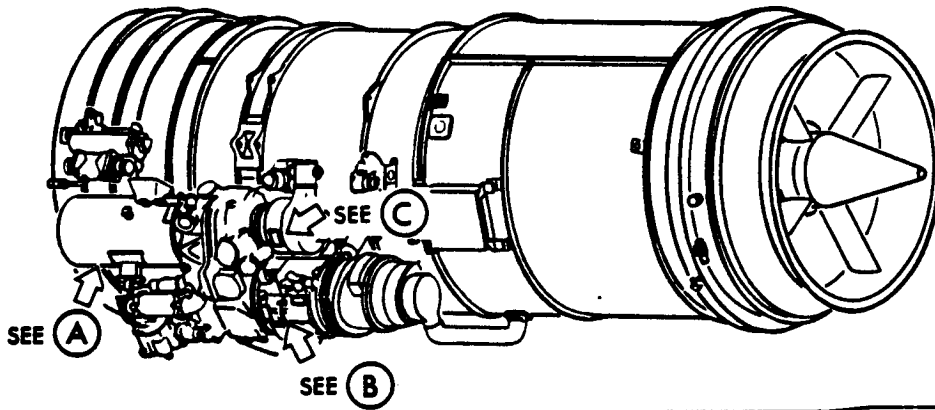
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	NOMENCLATURE							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1141-	69-50911-1		HOOK INSTL, LH COWL PANEL							A	RF
	69-50911-2		HOOK INSTL, LH COWL PANEL							B	RF
1	NAS1103-2		. BOLT								1
2	AN960C10L		. WASHER								2
3	NAS679A3W		. NUT								1
4	69-50910-1		. HOOK ASSY							A	1
4	69-50910-4		. HOOK ASSY							B	1
5	69-50910-2		. . HOOK								1
6	69-50910-3		. . RETAINER							A	1
6	69-50910-5		. . RETAINER							B	1
7	69-50919-1		. . SPRING								1
8	NAS1103-15		. . BOLT								1
9	NAS1057W3-065		. . SPACER								1
10	AN960C10L		. . WASHER								2
11	NAS679A3W		. . NUT								1



Metalcal Installation
Figure 1142

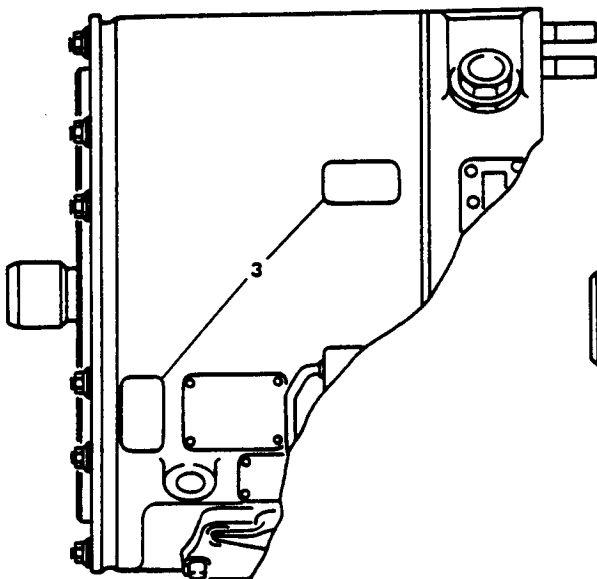
FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1142-1	69-60143-1 BAC27DPP-37										RF 1

OVERHAUL MANUAL



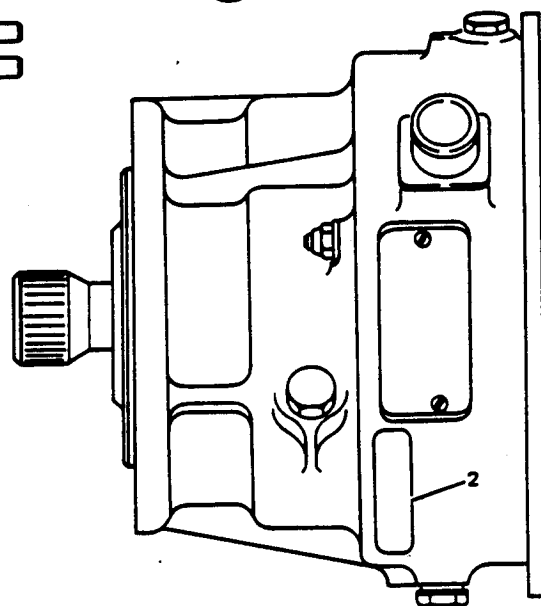
**ENGINE OIL TANK
LEFT SIDE VIEW**

(A)



**CONSTANT SPEED DRIVE
LEFT SIDE VIEW**

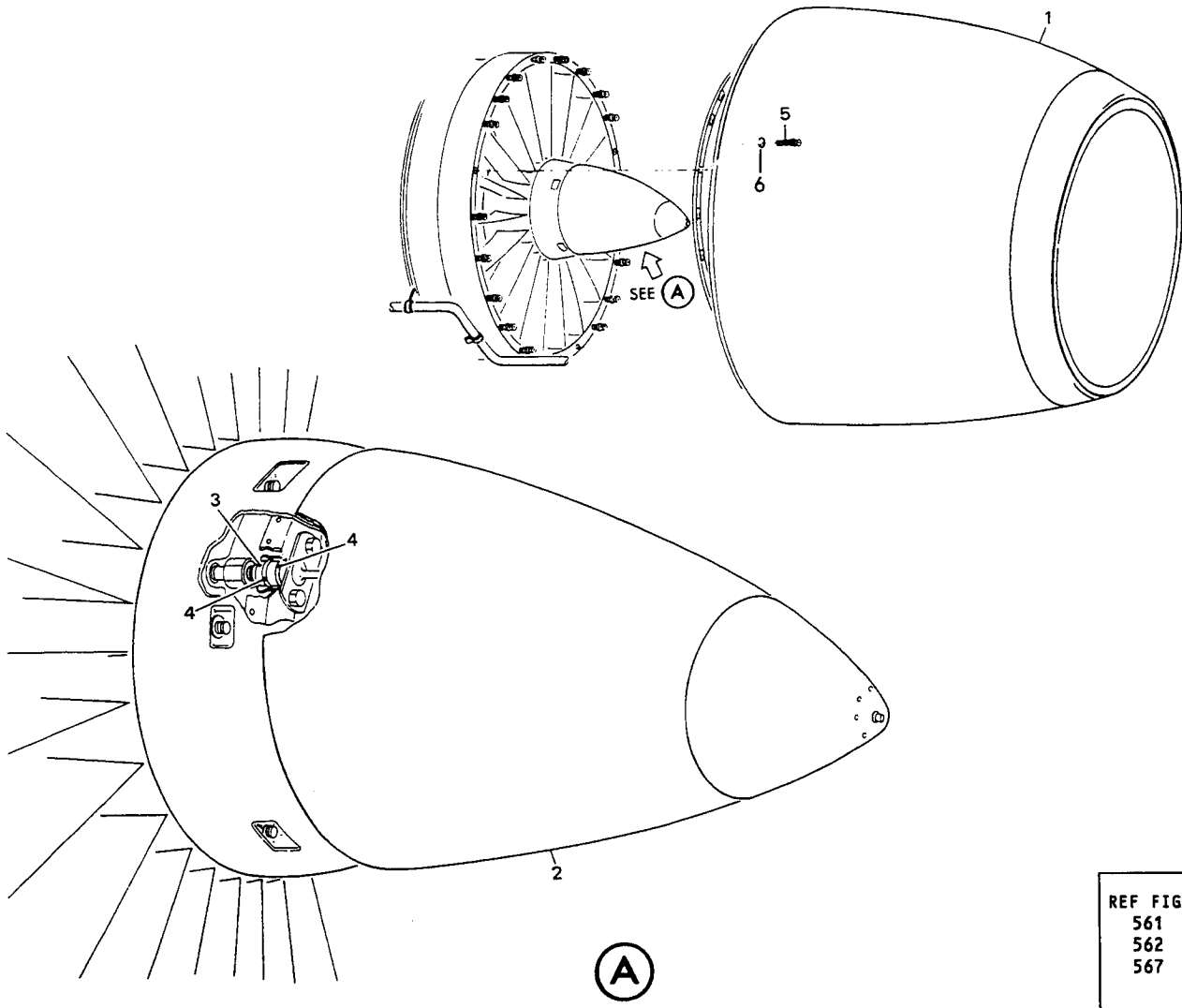
(B)



**STARTER
LEFT SIDE VIEW**

(C)

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1142A	65-69659-1		METALCAL	INSTL						A	RF
	65-69659-10		METALCAL	INSTL						B	RF
	65-69659-2		METALCAL	INSTL						C	RF
	65-69659-11		METALCAL	INSTL						D	RF
	65-69659-4		METALCAL	INSTL						E	RF
	65-69659-5		METALCAL	INSTL						F	RF
	65-69659-6		METALCAL	INSTL						G	RF
	65-69659-7		METALCAL	INSTL						H	RF
	65-69659-8		METALCAL	INSTL						I	RF
	65-69659-9		METALCAL	INSTL						J	RF
	65-69659-12		METALCAL	INSTL						K	RF
	65-69659-13		METALCAL	INSTL						L	RF
	65-69659-14		METALCAL	INSTL						M	RF
	65-69659-15		METALCAL	INSTL						N	RF
	65-69659-16		METALCAL	INSTL						O	RF
1	BACM10L6Z		.	METALCAL						A-KN	2
2	BACML10L1EAB		.	METALCAL						AEJN	2
2	BAC27DPP44		.	METALCAL						BD	2
2	BAC27DAP12		.	MARKER, FOIL						CDMO	2
2	BAC27DAP2		.	METALCAL						CJM	2
2	BACM10L6V		.	METALCAL						E	2
2	BACM10L1DXN		.	METALCAL						FI	2
2	BACM10L1IEKV		.	METALCAL						F	2
2	BACM10L6U		.	METALCAL						G	4
2	BAC27DPP35		.	METALCAL						H	4
2	BAC27DPP68		.	METALCAL						K	2
2	BAC27DPP70		.	METALCAL						L	2
2	BAC27DPP22		.	METALCAL						I	2
3	BACM10L1EBZ		.	METALCAL						AEN	AR
3	BAC27DPP45		.	METALCAL						BD	2
3	BAC27DAP1		.	METALCAL						CJM	2
3	BAC27DAP11		.	METALCAL						CDMO	AR
3	BACM10L6W		.	METALCAL						E	2
3	BACM10L1DXP		.	METALCAL						FI	2
3	BACM10L6Y		.	METALCAL						F	2
3	BACM10L6AB		.	METALCAL						G	2
3	BAC27DPP34		.	METALCAL						H	2
3	BAC27DPP67		.	METALCAL						K	2
3	BAC27DPP71		.	METALCAL						L	2
3	BAC27DPP23		.	METALCAL						I	2



Fat Lip Nose Cowl Installation
Figure 1143

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1143-	65-85379-1		FAT LIP NOSE COWL INSTL, ENGINE NO. 1 (VARIABLE)							A	RF
	65-85379-2		FAT LIP NOSE COWL INSTL, ENGINE NO. 2 (VARIABLE)							B	RF
	65-85379-3		FAT LIP NOSE COWL INSTL, ENGINE NO. 1 (BASIC)							C	RF
	65-85379-4		FAT LIP NOSE COWL INSTL, ENGINE NO. 2 (BASIC)							D	RF

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1143-	65-85379-5		FAT	LIP	NOSE	COWL	INSL	, ENGINE	E	RF	
	65-85379-6		FAT	LIP	NOSE	COWL	INSL	, ENGINE	F	RF	
1	65-85378-1		.	NOSE	COWL	ASSY,	NONACOUSTIC	*[2]	A	1	
								*[3] *[7] *[8]			
1	65-85378-2		.	NOSE	COWL	ASSY,	NONACOUSTIC	*[2]	B	1	
								*[3] *[7] *[8]			
1	65-85378-101		.	NOSE	COWL	ASSY,	ACOUSTIC (REPLD	BY 65-85378-311) *[2] *[4]	C	1	
								*[7] *[8]			
1	65-85378-311		.	NOSE	COWL	ASSY,	ACOUSTIC (REPLS	65-85378-101) *[2] *[5]	C	1	
								*[7] *[8] *[12]			
1	65-85378-102		.	NOSE	COWL	ASSY,	ACOUSTIC (REPLD	BY 65-85378-312) *[2] *[4]	D	1	
								*[7] *[8]			
1	65-85378-312		.	NOSE	COWL	ASSY,	ACOUSTIC (REPLS	65-85378-102) *[2] *[5]	D	1	
								*[7] *[8] *[12]			
1	65-86025-9		.	NOSE	COWL	ASSY	*[8] *[15] *[16]			1	
1	65-86025-10		.	NOSE	COWL	ASSY	*[8] *[15] *[16]			1	
1	65-85378-315		.	NOSE	COWL	ASSY,	ACOUSTIC *[2] *[6]	C	1		
								*[7] *[8] *[11]			
1	65-85378-316		.	NOSE	COWL	ASSY,	ACOUSTIC *[2] *[6]	D	1		
								*[7] *[8] *[11]			
1	65-85378-359		.	NOSE	COWL	ASSY,	NONACOUSTIC *[2]	A	1		
								*[3] *[7] *[8]			
1	65-85378-360		.	NOSE	COWL	ASSY,	NONACOUSTIC *[2]	B	1		
								*[3] *[7] *[8]			
1	65-85378-383		.	NOSE	COWL	ASSY	*[7] *[9] *[8] *[11]	C	1		
1	65-85378-383		.	NOSE	COWL	ASSY	*[7]	E	1		
1	65-85378-383		.	NOSE	COWL	ASSY	*[17]		1		
1	65-85378-384		.	NOSE	COWL	ASSY	*[7] *[9] *[8] *[11]	D	1		
1	65-85378-384		.	NOSE	COWL	ASSY	*[7]	F	1		
1	65-85378-384		.	NOSE	COWL	ASSY	*[17]		1		
1	65-85378-387		.	NOSE	COWL	ASSY	*[7] *[9] *[8] *[11]	C	1		
1	65-85378-387		.	NOSE	COWL	ASSY	*[17]		1		
1	65-85378-388		.	NOSE	COWL	ASSY	*[7] *[9] *[8] *[11]	D	1		
1	65-85378-388		.	NOSE	COWL	ASSY	*[17]		1		
1	65-85378-391		.	NOSE	COWL	ASSY	*[7] *[8]	A	1		
1	65-85378-392		.	NOSE	COWL	ASSY	*[7] *[8]	B	1		
1	65-85378-395		.	NOSE	COWL	ASSY	*[7] *[8] *[18]	C	1		
1	65-85378-395		.	NOSE	COWL	ASSY	*[7] *[13] *[17]		1		

OVERHAUL MANUAL

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY																		
			1	2	3	4	5	6	7																				
1143-																													
1	65-85378-396		.	N	O	S	E	C	O	W	L	A	S	S	*	[7]	*	[8]	*	[1	8]	D	1
1	65-85378-396		.	N	O	S	E	C	O	W	L	A	S	S	*	[7]	*	[1	3]						1
1	65-85378-396		.	N	O	S	E	C	O	W	L	A	S	S	*	[1	7]										1
1	65-85378-397		.	N	O	S	E	C	O	W	L	A	S	S															1
1	65-85378-398		.	N	O	S	E	C	O	W	L	A	S	S															1
2	65-85369-1		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-13		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-21		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-2		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-14		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-22		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-3		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-11		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-11		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-11		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-15		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-15		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-4		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-12		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-12		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-12		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-16		.	N	O	S	E	D	O	M	E	A	S	S															1
2	65-85369-16		.	N	O	S	E	D	O	M	E	A	S	S															1
3	69-18713-800		.	C	O	U	P	L	E	R																		1	
4	BACP11J113		.	O	-	R	I	N	G																			2	

FIG. & ITEM NO.	PART NO.	AIRLINE PART NUMBER	N O M E N C L A T U R E							USE CODE	QTY PER ASSY
			1	2	3	4	5	6	7		
1143-											
5	MS9148-08		.	B	O	L	T				23
6	AN960D516		.	W	A	S	H	E	R	CD	23
6	AN960D516		.	W	A	S	H	E	R	CD	23
6	AN960D516		.	W	A	S	H	E	R	ABEF	23

- *[1] REFER TO 71-13-15 FOR OVERHAUL.
- *[2] PRE SB 30-1010. NO REWORK OR EQUIVALENT NEW PART NO.
- *[3] POST SB 71-1045
- *[4] PRE SB 71-1033, 71-1131
- *[5] POST SB 71-1032, 71-1033
- *[6] POST SB 71-1063, 71-1064, 71-1065, 71-1092
- *[7] REFER TO 71-13-02 FOR OVERHAUL.
- *[8] LIMITED USAGE
- *[9] POST SB'S 71-1067, 71-1092, 71-1093
- *[10] POST SB'S 71-1032, 71-1033, 71-1055, 71-1063 THRU 71-1065, 71-1067, 71-1071, 71-1083, 71-1092, 71-1093, 71-1100, 71-1101, 71-1105, 71-1115
- *[11] POST SB'S 71-1071, 71-1083, 71-1092, 71-1100, 71-1101, 71-1105, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1142, 71-1147, 71-1151
- *[12] POST SB 71-1032
- *[13] POST SB 71-1115, 71-1118, 71-1121, 71-1125, 71-1125R1, 71-1125R2, 71-1128, 71-1140, 71-1142, 71-1147, 71-1151
- *[14] POST SB 71-1140
- *[15] POST SB 71-1131
- *[16] REWORKED FROM 65-85378-101 OR -102
- *[17] POST SB 71-1179, 71-1198
- *[18] POST SB 71-1278
- *[19] PREFERRED INSTALLATION CONFIGURATION

VENDORS

V06811 SYSTRON DONNER CORP., 200 SAN MIGUEL ROAD, CONCORD, CALIFORNIA
94520

V09049 CUSTOM COMPONENT SWITCHES, INC., 21111 PLUMMER STREET, CHATSWORTH,
CALIFORNIA 91311

V09268 TITFLEX DIVISION OF ATLAS CORP., PACIFIC DIVISION, 2978 NEBRASKA
AVENUE, SANTA MONICA, CALIFORNIA 90404

V09384 CONSOLIDATED ELECTRODYNAMICS CORP., TRANSDUCER DIVISION, 1400 S.
SHAMROCK, MONROVIA, CALIFORNIA 91016

V09402 GREATER MOUNTAIN CHEMICAL CO. INC., 15 JEREMY STREET, SALT LAKE
CITY, UTAH 84104

V10989 MECTRON INDUSTRIES, INC., 9857 REMER STREET SOUTH, EL MONTE,
CALIFORNIA 91733

V11328 TELEDYNE LINAIR ENGINEERING, 651, W KNOX STREET, GARDENA,
CALIFORNIA 90248

V11532 TELEDYNE RELAYS, 3155 WEST EL SEGUNDO BLVD., HAWTHORNE, CALIFORNIA
90250

V13636 BARRY WRIGHT CORP., 2323 VALLEY STREET, BURBANK, CALIFORNIA 91505

V14140 MCGRAW-EDISON CO., COMMERCIAL DEVELOPEMENT DIVISION, GRENIER FIELD
MUNICIPAL AIRPORT, MANCHESTER, NEW HAMSPHIRE 03103

V16519 METAL BELLOWS CORP., 20477 KNAPP STREET, CHATSWORTH, CALIFORNIA
91311

V16630 L.I. METAL INC., GLEN COVE, NEW YORK 11542

V18076 UMPCO INC., 12300 INDUSTRY STREET, GARDEN GROVE, CALIFORNIA 92645

V21465 SYSTRON-DONNER CORP., INTERNAL DIVISION, 1090 SAN MIGUEL RD.,
CONCORD, CALIFORNIA 94518

V25693 JOHN E. LINDBERG INC., 611 ADDISON STREET, BERKELEY, CALIFORNIA
94710

V29372 TRIDAIR FASTENERS, 3000 W. LOMETA BLVD, TORRANCE, CALIFORNIA
90505-5103

VENDORS

V29834 GENERAL ELECTRIC CO., INSTRUMENT DEPT., 40 FEDERAL STREET, WEST LYNN,
MASSACHUSETTS 01905

V33525 KIDDE, WALTER AND CO. INC., BELLEVILLE, NEW JERSEY 07109

V35918 LEWIS ENGINEERING COMPANY, 238 WATER STREET, NAUGATUCK, CONNECTICUT
06770

V44389 OAKITE PRODUCTS INC., 50 VALLEY ROAD, BERKELEY HEIGHTS, NEW JERSEY
07922

V50599 RESISTOFLEX CORPORATION, ROSELAND, NEW JERSEY 07068

V51057 VIBRO-METER CORPORATION, 22109 S. VERMONT AVE., TORRANCE, CALIFORNIA
90502

V51563 ROHR CORPORATION, CHULA VISTA, CALIFORNIA 92012

V54527 SHELL OIL COMPANY, 50 W. 50TH STREET, NEW YORK, NEW YORK 10020

V56232 SPERRY RAND CORPORATION, SPERRY GYROSCOPE DIVISION, GREAT NECK, NEW
YORK 11020

V59595 TEXACO INC., 135 E. 42ND STREET, NEW YORK, NEW YORK 10017

V59875 TRW INC AIRCRAFT COMPONENTS GROUP, POWER ACCESSORIES DIV., 23555
EUCLID AVE, CLEVELAND, OHIO 44117

V61102 TURCO PRODUCTS INC., 24600 SOUTH MAIN STREET, CARSON, CALIFORNIA 90745

V61349 AMETEK INC., UNITED STATES GAUGE DIVISION, SELLERSVILLE, PENNSYLVANIA
18960

V62983 VICKERS AOM DIVISION OF SPERRY RAND CORP., P.O. BOX 302, TROY,
MICHIGAN 48084

V70210 AIRESEARCH MFG. DIVISION OF GARRET CORPORATION, 2525 WEST 190 STREET
TORANCE, CALIFORNIA 90509

V72962 ELASTIC STOP NUT CORPORATION OF AMERICA, 2330 VAUXHALL ROAD, UNION,
NEW JERSEY 07083

V73760 I.T.T. GENERAL CONTROLS INC., 1200 SOUTH FLOWER STREET, BURBANK,
CALIFORNIA 91502

V75250 ABEX CORP., AEROSPACE DIVISION, 3151 WEST 5TH STREET, OXNARD,
CALIFORNIA 93032

OVERHAUL MANUAL

- V75361 KOEHLER-DAYTON INC., DIVISION OF LITTON INDUSTRIES, P.O. BOX 309,
SOUTH STREET, NEW BRITAIN, CONNECTICUT 06050
- V76005 LORD MFG. CO., DIVISION OF LORD CORP., 1635 W. 12TH STREET, ERIE,
PENNSYLVANIA 16512
- V76381 MINNESOTA MINING AND MFG. CO., ST. PAUL, MINNESOTA 55101
- V76541 MONSANTO CHEMICAL COMPANY, 800 NORTH LINDBERGH BLVD., ST. LOUIS,
MISSOURI 63166
- V77445 UNITED AIRCRAFT CORPORATION, PRATT & WHITNEY AIRCRAFT DIVISION,
400 MAIN, EAST HARTFORD, CONNECTICUT 06108
- V77932 SIMMONDS PRECISION PRODUCTS, INC., 105 WHITE PLAINS ROAD,
TARRYTOWN, NEW YORK 10591
- V78570 TITFLEX DIVISION OF ATLAS CORP., 603 HENDEE STREET, SPRINGFIELD,
MAINE 01109
- V78710 TELEFLEX, INC., P.O. BOX 218, NORTH WALES, PENNSYLVANIA 19454
- V79318 WHITTAKER CONTROLS, DIV WHITTAKER CORP, 12838 SATICOY ST., NORTH
HOLLYWOOD, CALIFORNIA 91605
- V79470 THE WEATHERHEAD COMPANY, 300 EAST 131 STREET, CLEVELAND, OHIO 44108
- V80234 VAPOR CORPORATION, 6420 WEST HOWARD STREET, CHICAGO, ILLINOIS 60648
- V81321 PUROLATOR PRODUCTS INC., 970 NEW BRUNSWICK AVENUE, RAHWAY, NEW
JERSEY 07065
- V81481 AVICA CORPORATION, P.O. BOX 149, NEWPORT, RHODE ISLAND
- V81873 HYDRAULIC RESEARCH, DIVISION OF TEXTRON INC., 25200 RYE CANYON
ROAD, VALENCIA, CALIFORNIA 91355
- V82829 ALLEN AIRCRAFT PRODUCTS IND., 4879 NEWTON FALLS, RAVENNA, OHIO
44266
- V83533 ESSEX CRYOGENICS INDUSTRIES, INC., 8213 GRAVOIS AVE., ST. LOUIS,
MISSOURI 63123
- V83843 WESTINGHOUSE ELECTRIC CORPORATION, AEROSPACE ELECTRICAL DIVISION,
P.O. BOX 989, LIMA, OHIO 45802
- V84971 TA MFG. CORPORATION, 4607 ALGER STREET, LOS ANGELES, CALIFORNIA
90039
- V86020 OZONE METAL PRODUCTS CORPORATION, 101-32 101ST STREET, OZONE PARK,
NEW YORK 11416
- V88334 THE WEATHERHEAD COMPANY, 8140 WEBB AVE., NORTH HOLLYWOOD,
CALIFORNIA 91605

OVERHAUL MANUAL

VENDORS

V88797 ROBINTECH INC., ELECTRO-MECHANICAL DIVISION, 3421 VESTAL ROAD,
 VESTAL, NY 13850

V89513 JANITROL DIVISION OF MIDLAND-ROSS CORP., 4200 SURFACE ROAD,
 COLUMBUS, OHIO 43228

V89944 KOLLSMAN INSTRUMENT DIVISION OF SUN CHEMICAL CORP., SYOSSET, NY

V90005 THE BENDIX CORPORATION, FILTER DIVISION, 434 WEST TWELVE MILE ROAD,
 MADISON HEIGHTS, MICHIGAN 48071

V92003 PARKER-HANNIFIN CORP., JAMBOREE BLVD., IRVINE, CALIFORNIA 92664

V92764 JOHNS-MANVILLE PRODUCTS CORP., 601 NASSAU STREET, NORTH BRUNSWICK,
 NEW JERSEY 08902

V97484 TECHNICAL DEVELOPMENT CO., 24 EAST GLENOLDEN AVENUE, GLENOLDEN,
 PENNSYLVANIA 19036

V97896 SUNDSTRAND DATA CONTROL INC., OVERLAKE INDUSTRIAL PARK, REDMOND,
 WASHINGTON 98052

V98441 STRATOFLEX INC., 220 ROBERTS CUT-OFF, FORT WORTH, TEXAS 76114

V98505 HYDRA ELECTRIC, 3151 KENWOOD STREET, BURBANK, CALIFORNIA 91503

V99167 SUNDSTRAND AVIATION DIVISION OF SUNDSTRAND CORPORATION, 4747
 HARRISON AVENUE, ROCKFORD, ILLINOIS 61101

V99193 AIRESEARCH MFG. CO. OF ARIZONA, 402 S. 36TH STREET, PHOENIX,
 ARIZONA 85034

V99755 FLEXIBLE METAL HOSE MFG. CO., 777 W. 16TH, COSTA MESA, CALIFORNIA
 92627

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
AE702087-2	1126-27	1	AN924-16		AR
AE702088-1	1126-25	1	AN924-4		AR
AE702089-1	1126-26	1	AN924-4J		AR
AE702120-10	1127-3	1	AN924-6		AR
AE702120-13	1127-4	1	AN924-6J		AR
AE702120-16	1127-5	1	AN92408		AR
AE702120	1127-6	1	AN924-8J		AR
AE702120-20	1127-7	1	AN924-10		AR
AE702120-29	1127-8	1	AN929A3		AR
AN122582		AR	AN960-10		AR
AN3C4A		AR	AN96010-L		AR
AN315-4R		AR	AN960-10L		AR
AN316-4R		AR	AN960-1216		AR
AN316C5R		AR	AN960-1216L		AR
AN320-4		AR	AN960-1416		AR
AN320C9		AR	AN960-416		AR
AN381-4-16	1101-11	1	AN960-416L		AR
AN4C14A		AR	AN960-6		AR
AN4045-1		AR	AN960-6L		AR
AN4047-1		AR	AN960-616L		AR
AN5C10A		AR	AN960C616L		AR
AN5C11A		AR	AN960-716		AR
AN500AD6-8		AR	AN960-916		AR
AN500AD8-7		AR	AN960-916L		AR
AN501AD10-7		AR	AN960C10		AR
AN501AD10-8		AR	AN960C10L		AR
AN6C11		AR	AN960C11	1126-14B	2
AN6289-10		AR	AN960C1216		AR
AN6289-12		AR	AN960C1216L		AR
AN6289-4		AR	AN960C416		AR
AN6289-6		AR	AN960C416L		AR
AN6289-8		AR	AN960C516		AR
AN735D56		AR	AN960C516L		AR
AN743P12		AR	AN960C616		AR
AN743BP12	1109-171A	1	AN960C616	1101-57	2
AN814-6CL		AR	AN960C916		AR
AN814-6SL		AR	AN960D10		AR
AN815-16D		AR	AN960D10L		AR
AN901-10C		AR	AN960D416		AR
AN901-24C		AR	AN960D416L		AR
AN901-4C		AR	AN960D516	1143-6	23
AN901-6C		AR	AN960D616		AR
AN901-8		AR	AN960PD10L	1117-61	AR
AN901-8C		AR	AN960PD416		AR
AN919-26D		AR	AN960PD416L		AR
AN919-28J		AR	AS136-06-0131	1125-35A	1
AN924-10		AR	AS136-06-0260	1125-16	1

737

BOEING
COMMERCIAL JET

POWERPLANT BUILDUP

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty. per Assy	Part No.	Fig. and Index No.	Qty. per Assy
AS137-04-0087	1125-15A	1	BACB3ONE5-10	1110-12	1
AS137-06-0146	1125-35A	1	BACB3ONE5-11	1110-12	1
AS138-04-0084	1125-15	1	BACB3ONE5H1	1130-25	1
AS138-04-0132	1127-9	1	BACB3ONE5H1	1133-24	1
AS138-06-0235	1127-17	1	BACB3ONF10D51	1101-18	1
AS138-08-0134	1127-18	1	BACB3NF5D11	1110-12	1
AS66651L8	1133-15	1	BACB3ONF5H1	1130-24	1
AS66651L8A	1133-15	1	BACCB1OBH12RW	1117-33	AR
AV18B1037-1	1125-50	1	BACCB1OBH14RW	1117-34	AR
BACB10A180D	1113-21	2	BACCB1OBH15RW	1117-35	AR
BACB10A187	1113-23	2	BACCB1OBH2RW	1117-25A	AR
BACB10A192M2	1112-45	1	BACCB1OBH3RW	1117-26	AR
BACB10J12	1112-23	1	BACCB1OBH4RW	1117-27	AR
BACB10J12	1112-44	1	BACCB1OBH5RW	1117-28	AR
BACB10J3	1112-9	1	BACCB1OBH6RW	1117-29	AR
BACB28X3C010	1112-12	2	BACCB1OBH7RW	1117-30	AR
BACB28X30C010	1112-26	2	BACCB1OBH8RW	1117-31	AR
BACB30CW4-4	1113-7	2	BACCB1OBH9RW	1117-32	AR
BACB30CW4H2	1113-4	2	BACCB1OBN362LR	1113-34	4
BACB30EJ4H2	1111-9	3	BACC10CD32	1125-89	2
BACB30EJ4H3	1111-3	12	BACC10DDC10	1120-16	3
BACB30EJ4H3	1111-14	1	BACC10DDC10	1126-29	1
BACB30EJ4H7	1111-7	2	BACC10DDC11	1127-10	7
BACB30FL3-2	1122-11	8	BACC10DDC12	1119-7	5
BACB30FM6A3U	1140-6	4	BACC10DDC12	1120-15	1
BACB30GU12-64	1101-2	2	BACC10DDC12	1126-28	1
BACB30LE5U6	1138-7	42	BACC10DDC12	1127-10	7
BACB30LE5U6	1138-12	32	BACC10DDC13	1127-11	8
BACB30LE5U6	1138-17	6	BACC10DDC16	1126-2	2
BACB30LE5U9	1138-12	84	BACC10DDC16	1127-12	6
BACB30LM3U2	1138-28	6	BACC10DDC20	1125-4	1
BACB30LM3U3	1138-31	4	BACC10DDC20	1125-4	2
BACB30LM3U3	1138-40	3	BACC10DDC20	1125-46	1
BACB30LM4U3	1138-33	4	BACC10DDC21	1127-12	7
BACB30LM5U10	1125-11A	2	BACC10DDC23	1125-8B	1
BACB30LM5U12	1125-11B	2	BACC10DDC4	1125-37	AR
BACB30LM5U9	1124-7	2	BACC10DDC4	1126-14	21
BACB30LU3-2	1122-11	1	BACC10DDC4	1126-30	2
BACB30LU3-6	1122-9	1	BACC10DDC4	1127-13	7
BACB3ONE3-3	1109-3D	4	BACC10DDC6	1119-13	1
BACB3ONE4-1	1128-23	2	BACC10DDC6	1119-17	1
BACB3ONE4-11	1110-7	1	BACC10DDC6	1119-31	1
BACB3ONE4D11	1110-7	1	BACC10DDC6	1119-35	2
BACB3ONF4-13	1109-115	6	BACC10DDC6	1120-14	3
BACB3ONF4-40	1118-2A	2	BACC10DDC6	1125-36	AR
			BACC10DDC6	1138-38	3

Part No.	Fig. and Index No.	Qty. per Assy.
BACC10DDC8	1120-14	3
BACC10DDC8	1138-39	3
BACC10DK2	1117-54	1
BACC10DP300A	1103-1	1
BACC10DP300A	1112-55	1
BACC10DU125C	1125-5	1
BACC10DU150C	1125-6	1
BACC10DU200X	1125-79	1
BACC10DU300C	1111-35	2
BACC10DU300Y	1111-35	2
BACC10DU350C	1111-22	4
BACC10DU350Y	1111-22	4
BACC10DU350C	1112-19	1
BACC10DU350Y	1112-19	1
BACC10DU350C	1112-33	1
BACC10DU350Y	1112-33	1
BACC10DU350C	1112-50	1
BACC10DU350Y	1112-50	1
BACC10DU350C	1112-57	2
BACC10DU350Y	1112-57	2
BACC10DU350C	1112-77	1
BACC10DU350Y	1112-77	1
BACC10DU350C	1113-35	1
BACC10DU350Y	1113-35	1
BACC10EP10	1120-16	3
BACC10EP10	1126-29	1
BACC10EP12	1119-7	5
BACC10EP12	1120-15	1
BACC10EP12	1126-28	1
BACC10EP12	1127-10	7
BACC10EP13	1127-11	8
BACC10EP14	1119-36	1
BACC10EP14	1127-11	8
BACC10EP16	1126-2	2
BACC10EP16	1127-12	6
BACC10EP17	1127-7B	1
BACC10EP18	1125-65	1
BACC10EP20	1125-4	3
BACC10EP20	1125-4	2
BACC10EP20	1125-46	1
BACC10EP21	1127-12	7
BACC10EP24	1125-73	1
BACC10EP24	1125-80	1
BACC10EP32	1125-84	1
BACC10EP4	1117-52	1
BACC10EP4	1125-37	AR
BACC10EP4	1126-14	1
BACC10EP4	1126-14	20
BACC10EP4	1126-30	2
BACC10EP4	1127-13	7

Part No.	Fig. and Index No.	Qty. per Assy.
BACC10EP5	1126-14	1
BACC10EP5	1117-51	1
BACC10EP6	1103-21	3
BACC10EP6	1119-31	1
BACC10EP6	1119-35	2
BACC10EP6	1119-45	12
BACC10EP6	1125-36	AR
BACC10EP6	1125-84	1
BACC10EP6	1126-62	2
BACC10EP6	1138-38	1
BACC10EP7	1138-27	6
BACC10EP8	1120-14	3
BACC10EP8	1138-39	3
BACC10EP9	1127-44	1
BACC10GE3C	1103-22	1
BACC10GE4	1103-34	1
BACC10GE4C	1103-23	1
BACC10GU102	1128-60	1
BACC10GY400	1103-17	1
BACC10HS()	1117-37A	7
BACC10HS()	1127-42	AR
BACC10HS04	1117-52	1
BACC10HS04	1126-14	AR
BACC10HS04	1126-30	2
BACC10HS04	1127-43	1
BACC10HS05	1117-51	1
BACC10HS05	1126-74	1
BACC10HS06	1103-21	3
BACC10HS06	1119-13	1
BACC10HS06	1119-17	1
BACC10HS06	1119-31	1
BACC10HS06	1119-35	2
BACC10HS06	1119-45	12
BACC10HS06	1125-36	AR
BACC10HS06	1125-37	AR
BACC10HS06	1125-85	1
BACC10HS06	1126-49A	3
BACC10HS06	1126-62	2
BACC10HS07	1127-44	1
BACC10HS08	1120-14	3
BACC10HS09	1127-44	1
BACC10HS10	1120-16	4
BACC10HS10	1120-16A	2
BACC10HS10	1126-29	1
BACC10HS11	1126-52G	2
BACC10HS12	1119-7	5
BACC10HS12	1120-15	1
BACC10HS12	1126-28	1
BACC10HS12	1127-10	7
BACC10HS13	1127-12	6

Part No.	Fig. and Index No.	Qty. per Assy.
BACC10HS14	1119-36	1
BACC10HS14	1127-11	8
BACC10HS16	1126-2	2
BACC10HS16	1127-12	6
BACC10HS17	1127-7B	1
BACC10HS18	1125-65	1
BACC10HS20	1103-32	1
BACC10HS20	1125-4	AR
BACC10HS20	1125-46	1
BACC10HS20	1126-49C	1
BACC10HS21	1127-12	6
BACC10HS23	1126-8B	1
BACC10HS24	1125-73	1
BACC10HS24	1125-80	1
BACC10HS32	1125-84	1
BACC10Y22-6C	1125-1	1
BACC12CB3	1117-36	AR
BACC14B1CH	1107-13	2
BACC14H4L	1126-18	2
BACC14H4LJ	1125-60	1
BACC14H8LS	1124-16	1
BACC3026	1140-8	4
BACD40D10-12-30	1122-35	1
BACD40D10-15-30	1122-36	1
BACD40D14-12-23	1113-33	2
BACD40D14-12-36	1113-33	2
BACE21AS0406	1130-17	1
BACE21AS0406	1130A-14	1
BACE21AS0406	1133-12	1
BACE21AW0606	1127-22	1
BACF22S32AC	1125-14	1
BACG10AG305	1126-5	1
BACG10U4	1131-2	1
BACG10U4	1115-4A	1
BACG10U4	1115-28	1
BACG21A10	1107-14	2
BACH6R0665DUW	1127-7	1
BACH6R0684DL-W	1127-7	1
BACH7D0137DD	1125-19	1
BACH30AS06F0201	1119-41	1
BACH30AS08F0173	1120-5	1
BACH30AS10F0173	1120-6	1
BACH30AU08F0116	1120-3	1
BACH30AV06F0163	1119-54	1

Part No.	Fig. and Index No.	Qty. per Assy.
BACH30AV08F0116	1120-3	1
BACH30AV10F0114	1120-4	1
BACH5P0184CCF	1119-41	1
BACH5R0173DDF	1120-5	1
BACH5R0173EEF	1120-5	1
BACH7D137DD	1125-19	1
BACJ40AD45-9	1128-26	1
BACJ40AD67-12	1128-16	2
BACJ40A20-5	1105-10	1
BACJ40A20-9	1105-13	1
BACJ4DA21-6	1122-27	1
BACJ40A21-8	1122-27	1
BACJ40AD45-6	1128-22M	1
BACJ40AD57-4	1128-21	1
BACJ40AD57-6	1128-22	1
BACJ40K7A6A6	1107-3	1
BACM10F11-3AH	1136-9	1
BACM10L01BJZ	1123-14	1
BACM10LDXN	1142A-2	2
BACM10L1DXP	1142A-3	2
BACM10L1EAB	1142A-2	2
BACM10L1EBZ	1142A-3	2
BACM10L6AB	1142A-3	2
BACM10L6U	1142A-2	2
BACM10L6V	1142A-2	4
BACM10L6W	1142A-3	2
BACM10L6Y	1142A-3	2
BACM10L6Z	1142A-1	1
BACM10P6GS	1132-14	1
BACM10P6GT	1132-15	1
BACM10P6GU	1123-15	1
BACN10BL4L	1113-30	6
BACN10BL4L	1113-3	2
BACN10BL4L	1113-8	4
BACN10BL6L	1103-6	5
BACN10BL6L	1103-6	5
BACN10BL6L	1114-16	12
BACN10BY512	1101-5	2
BACN10DR6J	1103-19	1
BACN10DU6	1103-15	1
BACN10GW3A	1138-6	1
BACN10GW6	1103-6	5
BACN10GW6	1114-16	12
BACN10HR5C	1138-10	42

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty. per Assy	Part No.	Fig. and Index No.	Qty. per Assy
BACN10HR5C	1138-15	32	BACP11PB12	1119-6	1
BACN10HR5CD	1101-59	1	BACP11PB12	1123-6	2
BACN10HX16	1127-35A	2	BACP11PB4	1111-49	1
BACN10JC	1122-13	8	BACP11PB4	1115-4	2
BACN10JC12	1101-5	2	BACP11PB4	1115-4	1
BACN10JC3	1109-3E	4	BACP11PB4	1115-23	1
BACN10JC3	1115-8	4	BACP11PB4	1115-23	2
BACN10JC3	1115-16	1	BACP11PB4A	1115-28	1
BACN10JC3	1122-24	1	BACP11PB4	1118-6	2
BACN10JC3	1125-75	1	BACP11PB4	1125-24	1
BACN10JC3	1125-82	1	BACP11PB4	1126-21	1
BACN10JC3	1125-87	1	BACP11PB4	1126-22	1
BACN10JC3	1125-91	4	BACP11PB4	1126-36H	2
BACN10JC3	1126-36B	2	BACP11PB4	1126-35	1
BACN10JC3	1127-7E	2	BACP11PB4	1126-41	2
BACN10JC3	1128-28	5	BACP11PB4	1126-48	2
BACN10JC3	1117-60	AR	BACP11PB4A	1115-4A	1
BACN10JC3	1138-30A	1	BACP11PB5A	1139-6	2
BACN10JC3C	1138-30	2	BACP11PB6	1111-37	1
BACN10JC3CM	1103-27	2	BACP11PB6	1111-45	1
BACN10JC4		AR	BACP11PB6	1119-12	1
BACN10JC4	1110-8	1	BACP11PB6	1119-16	1
BACN10JC4	1128-24	2	BACP11PB6	1119-21	2
BACN10JC4	1128-32	1	BACP11PB6	1119-34	1
BACN10JC4	1130-26	1	BACP11PB6	1119-40	1
BACN10JC4	1133-27	1	BACP11PB6	1123-11	1
BACN10JC5	1110-13	1	BACP11PB6	1125-25	1
BACN10JD104	1110-13	1	BACP11PB6	1125-52	1
BACN10JD105	1110-13	1	BACP11PB6	1126-32	1
BACN10JM60	1127-1A	2	BACP11PB6	1132-9	1
BACN10LHC38K	1112-73F	2	BACP11PB6A	1139-4	1
BACN10MK3-45	1109-169	1	BACP11PB8	1111-41	1
BACN10MK3-85	1109-106A	1	BACP11PB8	1111-51	1
BACN10T5A	1102-9	4	BACP11PB8	1114-14	1
BACP11J113	1102-12	2	BACP11PB8	1125-51	1
BACP11J113	1143-4	2	BACP11PB8	1132-10	1
BACP11K6	1103-12	1	BACP11PB8	1135-6	1
BACP11PB5A	1139-6	2	BACP11PB8A	1139-2	1
BACP11PB6	1103-3	1	BACP11P11	1115-24	2
BACP11PB6A	1139-6A	1	BACP11P11	1126-22	1
BACP11PB8	1124-20	1	BACP11P112	1114-6	2
BACP11PB8A	1139-6A	1	BACP11P21	1132-11	2
BACP11PB10	1114-10	1	BACP11P228	1107-4	1
BACP11PB10	1135-8	1	BACP11P26	1107-17	2
BACP11PB11	1126-42	2			

Part No.	Fig. and Index No.	Qty. per Assy.
BACP20AU8	1124-18	1
BACP20AU8J	1112-59A	1
BACP20AU8J	1126-64	1
BACR10G32	1109-42B	1
BACR10G41	1109-40	2
BACR10G41	1109-45	2
BACR10G63	1109-56	2
BACS12CB04	1117-59	AR
BACS12CB06-4	1115-4	4
BACS12CB06-6	1109-170	2
BACS12CB06-7	1115-9	1
BACS12CB3-(-)	1126-23A	AR
BACS12CB3-(-)	1127-31	AR
BACS12CB3-10	1124-22	2
BACS12CB3-12	1125-43	1
BACS12CB3-14	1117-8	2
BACS12CB3-16	1125-48	1
BACS12CB3-20	1119-45C	2
BACS12CB3-22	1117-47	1
BACS12CB3-24	1117-45	1
BACS12CB3-26	1120-18	1
BACS12CB3-32	1117-44	2
BACS12CB3-48	1117-6	1
BACS12CB3-5	1115-13	1
BACS12CB3-5	1117-12A	4
BACS12CB3-6	1105-11	1
BACS12CB3-6	1109-92	2
BACS12CB3-6	1109-100	2
BACS12CB3-6	1109-102B	2
BACS12CB3-6	1109-153	2
BACS12CB3-6	1109-157	4
BACS12CB3-6	1109-181	2
BACS12CB3-6	1109-201	2
BACS12CB3-6	1109-217	4
BACS12CB3-6	1109-92	2
BACS12CB3-6	1113-43	1
BACS12CB3-6	1116-3	3
BACS12CB3-6	1119-35A	2
BACS12CB3-6	1119-60	1
BACS12CB3-6	1125-74	1
BACS12CB3-6	1125-81	1

Part No.	Fig. and Index No.	Qty. per Assy.
BACS12CB3-6	1125-86	1
BACS12CB3-6	1125-90	4
BACS12CB3-6	1128-10	8
BACS12CB3-6	1128-5	4
BACS12CB3-6	1128-5	1
BACS12CB3-7	1105-14	1
BACS12CB3-7	1109-148	6
BACS12CB3-7	1109-161	5
BACS12CB3-7	1109-171B	1
BACS12CB3-7	1109-187	4
BACS12CB3-7	1109-210	4
BACS12CB3-7	1109-220	4
BACS12CB3-7	1115-18	1
BACS12CB3-7	1116-8	2
BACS12CB3-7	1117-7	2
BACS12CB3-7	1117-36	120
BACS12CB3-7	1117-55	1
BACS12CB3-7	1119-13A	1
BACS12CB3-7	1119-17A	1
BACS12CB3-7	1119-31A	1
BACS12CB3-7	1119-35A	2
BACS12CB3-7	1119-45B	11
BACS12CB3-7	1125-39	7
BACS12CB3-7	1125-44	1
BACS12CB3-7	1125-66	1
BACS12CB3-7	1126-29A	1
BACS12CB3-7	1126-36A	2
BACS12CB3-7	1126-63	2
BACS12CB3-7	1126-8A	2
BACS12CB3-7	1126-13A	1
BACS12CB3-7	1127-29	9
BACS12CB3-7	1128-5	4
BACS12CB3-8	1109-191	1
BACS12CB3-8	1109-195	1
BACS12CB3-8	1112-79	2
BACS12CB3-8	1115-13	1
BACS12CB3-8	1116-14	2
BACS12CB3-8	1117-41	3
BACS12CB3-8	1118-2	2
BACS12CB3-8	1119-7A	4

Part No.	Fig. and Index No.	Qty. per Assy	Part No.	Fig. and Index No.	Qty. per Assy
BACSL2CB3-8	1120-19	2	BL1705-5-1	1115-5	1
BACSL2CB3-8	1124-11	2	B5069110302-0	1104-1	2
BACSL2CB3-8	1127-7C	2	B520BBB0101-000	1126-25	1
BACSL2CB3-9	1105-15	1	B520DDB0104-000	1126-27	1
BACSL2CB3-9	1109-12	3	B520DDD0101-287	1126-26	1
BACSL2CB3-9	1115-6	4	B6002	1139-3	1
BACSL2CB3-9	1117-20	2	B6003	1139-1	1
BACSL2CB3-9	1117-24	2	B7670	1139-5	2
BACSL2CB3-9	1125-38	5			
BACSL3AP4	1127-2A	2	C81662-2	1138-21	1
BACSL3AP6	1127-1C	2			
BACSL3BD60	1127-1B	2	D58364	1113-15	1
BACSL8AN12	1126-23C	1	D9798	1126-31	1
BACS40L12-28	1111-25	1			
BACT12C20	1117-48	2	ER10339	1127-15	1
BACT16AX060406	1125-31	1	ER1832-6	1127-22A	1
BACU24AA0406	1126-17	1	F6100236M3	1125-71	1
BACW10AK-WP5	1109-133	1	F6100236M4	1125-71	1
BACW10BP4CT	1111-4	12	F61D0236M6	1125-71	1
BACW10BP4CT	1111-10	6			
BACW10BP4CT	1111-16	6	GA7850	1126-17	1
BACWH101	1107-15	4			
BACW10BP3APU	1103-10	2	K710-22	1115-5	1
BACW10BP5ACU	1138-8	42			
BACW10BP5ACU	1138-13	32	LER1832-6	1127-22A	1
BACW10P16S	1109-134	1			
BACW10P186C			MODEL 750	1106-4	2
BACW10P16S	1109-134	1	MODEL 9145	1106-4	2
BACW10P186C	1140-7	2			
BAC27DAP1	1142A-3	2	MS20002C12		AR
BAC27DAP11	1142A-2	2	MS20002C4		AR
MS20002C5	1101-45	AR	MS20426D3-(-)		AR
BAC27DAP12	1142A-2	2	MS20426D3-4		AR
BAC27DAP2	1142A-2	2	MS20426D3-5		AR
BAC27DPP22	1142A-2	2	MS20427M3	1112-73G	4
BAC27DPP23	1142A-3	2	MS20615-4M		AR
BAC27DPP34	1142A-3	2	MS20756-24		AR
BAC27DPP35	1142A-2	4	MS21042L3	1125-2A	2
BAC27DPP37	1142-1	1	MS21902-10		AR
BAC27DPP44	1142A-2	2	MS21902-12		AR
BAC27DPP45	1142A-3	2	MS21902-16J		AR
BAC27DPP67	1142A-3	2	MS21902-4		AR
BAC27DPP68	1142A-2	2	MS21902-4C		AR
BAC27DPP70	1142A-2	2	MS21902-4J		AR
BAC27DPP71	1142A-3	2	MS21902-6		AR

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
MS21902-6J		AR	MS35338-41		AR
MS21902-8		AR	MS35338-43		AR
MS21902-8J		AR	MS9015-04		AR
MS21907-4		AR	MS9020-20		AR
MS21907J4	1125-30A	1	MS9021-021	1108-2	AR
MS21908-10		AR	MS9034-08		8
MS21908-12		AR	MS9034-10		AR
MS21908-4		AR	MS9058-20		AR
MS21908-6		AR	MS9099-20		AR
MS21908-8		AR	MS9147-18		AR
MS21909-6		AR	MS9147-22		AR
MS21912-4		AR	MS9148-08		AR
MS21912-4C		AR	MS9386-161	1113-27	1
MS21912-4J		AR	MS9387-04		AR
MS21912-6		AR			
MS21916J6-4	1126-36I	1	NAS1022A12		AR
MS21916J6-4	1126-49	2	NAS1057W3-065		AR
MS21916-10-8		AR	NAS1067A3		AR
MS21916-12-8		AR	NAS1090-1		AR
MS21916-6-4		AR	NAS1056E6-031		AR
MS21921-4	1127-2B	2	NAS1103-1		AR
MS21921-6	1127-1D	2	NAS1103-15		AR
MS21924-10		AR	NAS1103-2		AR
MS21924-4		AR	NAS1103-3		AR
MS21924-4J		AR	NAS1103-4		AR
MS21924-6		AR	NAS1103-6		AR
MS21926-4		AR	NAS1103-7		AR
MS21926-6		AR	NAS1103-8		AR
MS24665-138		AR	NAS1104-11		AR
MS24665-370		AR	NAS1104-12		AR
MS24665-88	1101-64	1	NAS1104-13		AR
MS24673-3		AR	NAS1104-15		AR
MS25083-2AB5		AR	NAS1104-16		AR
MS25083-2BB10		AR	NAS1104-18		AR
MS25083-2BB4		AR	NAS1104-3		AR
MS25083-28CB		AR	NAS1104-3W	1132-4A	4
MS26777-10		AR	NAS1104-30		AR
MS2677-12		AR	NAS1104-36		AR
MS27294-3	1117-58	1	NAS1104-4		AR
MS28777-4		AR	NAS1104-5		AR
MS28777-6		AR	NAS1104-6		AR
MS28777-8		AR	NAS1104-7		AR
MS29512-20		AR	NAS1104-8		AR
MS29513-21		AR	NAS1104-9		AR
MS29561-228		AR			

Part No.	Fig. and Index No.	Qty. per Assy.
NAS1105-10		AR
NAS1105-3		AR
NAS1201CA8A		AR
NAS1236-4H		AR
NAS1237-4		AR
NAS1303-1		AR
NAS1303-1H		AR
NAS1303-2		AR
NAS1304-16H		AR
NAS1304-2		AR
NAS1304-2H		AR
NAS1304-3		AR
NAS1304-4H		AR
NAS1304-9		AR
NAS1305-4		AR
NAS1306-2		AR
NAS1454C4-0208		AR
NAS1611-113		AR
NAS1612-10		AR
NAS1612-16		AR
NAS1612-20		AR
NAS1612-4		AR
NAS1612-6		AR
NAS1716C20M	1125-1A	1
NAS1801-3-14		AR
NAS1801-3-30	1103-36	1
NAS1801-3-6		AR
NAS1801-3-7	1128-5M	3
NAS1802-307		AR
NAS1802-308		AR
NAS1802-3-10		AR
NAS1802-3-8	1125-1B	2
NAS1804-6	1101-56	2
NAS43DD3-128		AR
NAS43DD4-16		AR
NAS43DD4-26		AR
NAS43DD4-28		AR
NAS43HT3-11		AR
NAS43HT3-16		AR
NAS43HT3-20		AR
NAS43HT3-24		AR
NAS43HT3-32		AR
NAS43HT3-40		AR
NAS43HT3-48		AR
NAS43HT3-56		AR
NAS43HT3-64		AR
NAS43HT3-8		AR
NAS43HT3-80	1103-35	1
NAS43T4-11		AR
NAS514P440-5		AR

Part No.	Fig. and Index No.	Qty. per Assy.
NAS514P-1032-16	1117-13J	2
NAS517-2-1		AR
NAS551-4H		AR
NAS566-39	1117-38B	1
NAS595-20E		AR
NAS595-24E		AR
NAS623-3-3		AR
NAS623-3-4		AR
NAS6604-1		AR
NAS6606-10	1101-58	2
NAS6712EG4	1101-2	2
NAS679A04W		AR
NAS679A06W		AR
NAS679A3		AR
NAS679A3	1128-6M	3
NAS679A3W		AR
NAS679A3W	1128-6	1
NAS679A4		AR
NAS679A4W		AR
NAS679A5		AR
NAS679A6		AR
NAS679A6W		AR
NAS679C3M	1126-14D	2
NAS679C3W		AR
NAS679C4W		AR
NAS679C5		AR
NAS76A4-005P		AR
OMP2505-2	1132-8	1
OMP2505-4	1132-13	1
OMP2506-2	1114-4	1
OMP2506-2	1123-12	1
OMP2506-2	1132-7	1
OMP2506-2	1132-8	1
OMP2506-4	1114-5	1
OMP2506-4	1123-13	1
OMP2506-4	1132-12	1
OMP2506-4	1132-13	1
R18423-1A	1101-1	2
R18424-1A	1101-6	1
ST104AB	1115-1	1
ST104Z	1115-1	1
S387SSM32	1125-89	2
S41B90041060105	1126-27	1
S41B90042040102	1126-25	1
S41B90043060101	1126-26	1
TA12100004	1109-152	1
TA12100004	1109-180	1

Part No.	Fig. and Index No.	Qty. per Assy.	Part No.	Fig. and Index No.	Qty. per Assy.
TA12100004	1128-13	2	10-60563-33	1127-6A	1
TA5000BH14AT	1128-8	2	10-60563-34	1127-6A	1
TA5000BH16HB	1128-8	2	10-60563-35	1127-3A	1
TA5000BH8A	1109-152	1	10-60706-10	1103-2	1
TA5000BH8A	1109-180	1	10-60706-2	1103-2	1
TA5000BH8AT	1109-152	1	10-60706-3	1103-2	1
TA5000BH8AT	1109-180	1	10-60706-4	1103-2	1
TA5000BH8AT	1128-13	2	10-60706-5	1103-2	1
			10-60706-8	1103-2	1
WL750	1106-4	2	10-60722-4	1107-5	1
WL800	1106-4	2	10-60792-1	1124-1	1
			10-60817-7	1124-5	1
015067S2-16	1127-27	1	10-60817-8	1124-5	1
01567S5-16	1127-28	1	10-60927-3	1103-11	1
053034	1136-8	1	10-61066-10	1114A-2	1
			10-61095-35	1128-7	1
1G124	1103-11	1	10-61095-40	1128-7	1
1G309	1103-11	1	10-61096-105	1128-7	1
			10-61096-115	1128-1	1
1-01188	1126-31	1	10-61096-19	1128-12	1
1-01288	1126-31	1	10-61096-615	1128-1	1
			10-61096-9	1128-12	1
10-3269-8	1115-17	1	10-61223-2	1114-2	1
10-3269-8	1118-1	1	10-61224-2	1114A-3	1
10-60246-10	1133-15	1	10-61224-10	1114-1	1
10-60295-15	1114-3	1	10-61224-10	1114A-3	1
10-60470-10	1130A-15	1	10-61224-12	1114-1	1
10-60470-4	1130-18	1	10-61224-5	1117-1	1
10-60470-5	1130-18	1	10-61224-9	1114-1	1
10-60470-7	1130-18	1	10-61233-1	1135-2	1
10-60492-4	1111-34	1	10-61233-4	1135-1	1
10-60495-6	1112-56	1	10-61348-14	1138-21	1
10-60295-15	1114A-1	1	10-61348-4	1138-21A	1
10-60507-1	1105-1	1	10-61348-7	1138-21	1
10-60507-14	1105-1	1	10-61357-2	1126-25	1
10-60507-17	1105-1	1	10-61357-3	1126-26	1
10-60507-6	1105-1	1	10-61357-4	1126-27	1
10-60508-3	1111-21	1	10-61794-1	1133-15	1
10-60508-3	1112-32	1	10-61813-27	1138-11	1
10-60517-39	1101-1	2	10-61813-28	1138-11	1
10-60517-49	1101-1	2	10-61813-33	1138-11	1
10-60517-40	1101-6	1	10-61813-34	1138-11	1
10-60517-50	1101-6	1	10-61813-37	1138-11	1
10-60555-3	1136-8	1	10-61813-38	1138-11	1
10-60563-10	1127-3	1	10-61813-57	1138-11	1
10-60563-13	1127-4	1	10-61813-58	1138-11	1
10-60563-16	1127-5	1	10-61813-59	1138-11	1
10-60563-19	1127-6	1	10-61813-7	1138-11	1
10-60563-20	1127-7	1	10-61813-8	1138-11	1
10-60563-21	1127-8	1	10-61988-1	1110-20	1
10-60563-29	1127-8	1	101A052-3	1117-17B	2

Part No.	Fig. and Index No.	Qty per Assy
10646-3	1125-19	1
10647-3	1125-17	1
10648-3	1125-16	1
10649-3	1126-45	1
11-10865	1125-53	1
1112-589971	1124-15	1
118-10398	1126-31	1
118-10426	1125-29	1
118-10430	1125-49	1
118-10530	1126-60	1
118-10555	1119-19	1
118-10557	1126-39	2
118-10560	1123-10	1
123558-1-4	1111-21	1
129240-1-1	1112-64	1
129525	1124-5	1
144-904-000-01	1106-4	1
144-904-000-04	1106-4	1
144-904-000-05	1106-4	1
1510376-2	1106-1	2
1611-0716	1114-21	1
16228	1125-17	1
16229	1125-16	1
16230	1125-18	1
16231	1127-9	1
16233	1125-15	1
16236	1125-19	1
1667	1125-50	1
1705	1128-7	1
1714-600-5	1128-12	1
173-1010-2	1138-11	1
173-1010-501	1138-11	1
173-1010-503	1138-11	1
173-1010-509	1138-11	1
173-1010-511	1138-11	1
173-1010-517	1138-11	1
173-1010-519	1138-11	1
173-1010-523	1138-11	1
1784340-10	1106-3	2
182400-1-1	1113-13	1
2CM9ABY7	1104-1	2
2-01066	1127-15	1
2-01071	1126-17	1
2-01557	1103-13	1
2-01575	1126-39	2
2-02302	1119-19	1
2-02303	1123-10	1
20-18856	1125-7A	1
207865	1128-3	2

Part No.	Fig. and Index No.	Qty per Assy
207865	1128-18	2
209592	1117-12B	4
209592	1117-17A	4
214335-8	1109-151	1
22103	1125-53	1
225485	1124-5	1
226195	1105-7	1
23318-1	1138-21A	1
24096-150C	1125-70	2
24502-150	1125-69	2
25604	1138-21	1
26130105	1124-1	1
26130105-01	1124-1	1
2680337	1103-2	1
2730426	1103-2	1
2760000-101	1112-56	1
3-58452	1107-16	2
307002S2-6	1127-23	1
307002S2-8	1127-24	1
307002S5-6	1127-25	1
307002S5-8	1127-26	1
3163408	1112-65	1
320517	1133-19A	1
332A1025-2	1122-14	1
354117	1105-6	1
360527	1133-19F	1
372873	1105-24	4
374105	1133-19B	1
377129	1113-27	1
382624	1133-19H	1
383152-1-2	1103-5	1
383152-16-1	1103-5	1
383152-19-1	1103-5	1
383342-1-1	1103-5	1
387999	1133-19D	1
392664-1-1	1113-14	1
392686-3-1	1111-43	1
392688-1-1	1103-2	1
392688-5-1	1103-2	1
392708-1-1	1124-1	1
392716-1-1	1111-34	1
4-125-0001	1106-4	2
4-125-0001	1106-4A	1
4-126-001	1106-4	2
400922	1122-25	1
411-00117	1124-15	1
42D107A1	1115-17	1
42D107A1	1118-1	1
447884	1105-18	1

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
447885	1105-20	1	534093	1109-29E	1
447887	1105-5	1	540966	1109-104A	1
451094	1109-83	5	55097	1130-18	1
451094	1109-86	1	55098	1130-18	1
454423	1109-113	6	55098-01	1130A-15	1
454423	1109-123A	1	55589	1114A-21	1
454423	1109-124A	1	55745	1130-5	1
454425	1109-199	2	55759	1130-6	1
463647	1109-123	1	56B17C	1131-1	1
463649	1109-124	1	56B17D	1131-1	1
463663	1109-25	1	567326	1109-62	1
463712	1109-78	1	577994	1109-67	1
463749	1109-98	1	578270	1109-17	1
463749	1109-102	2	578366	1109-27	1
463717	1109-74	1	578494	1109-117	1
463749	1109-103	2	578837	1109-96	1
463974	1109-2A	7	581727	1109-208	1
463974	1109-3	10	581728	1109-207	1
463983	1109-6A	1	581733	1109-82	1
454425	1109-206	1	581734	1109-79	2
464464	1109-6B	1	581736	1109-75	1
466471	1109-47	1	581738	1109-76	1
466472	1109-32	1	581739	1109-77	1
466472	1109-48	1	581886	1109-4	1
466630	1109-59	2	581889	1109-36	1
466782	1109-97	1	581892	1109-42	1
467489	1109-37	1	581895	1109-35	2
468904	1109-142	1	581897	1109-34A	1
480389	1109-18	2	583198	1109-7	1
480389	1109-29D	1	583304	1109-15	1
490479	1109-60	1	583306	1109-16	1
490545	1109-29	1	583476	1109-68	1
490546	1109-28	1	583477	1109-64	1
491578	1109-65	1	583840	1109-20D	1
491585	1109-19	3	583842	1109-26	1
491585	1109-20	1	583845	1109-24	1
491585	1109-20A	1	584216	1109-120	1
491585	1109-22	2	584219	1109-118	1
491585	1109-23	1	584288	1109-61	1
491585	1109-23A	2	584289	1109-89	1
491631	1109-82E	1		1109-94	1
491641	1109-8	7	584294	1109-95	1
			584296	1109-85	1
501776	1109-8	7	584298	1109-104	1
511804	1109-198	1	584309	1109-183	1
511806	1109-197	3	584319	1109-163	1
	1109-198	1	584320	1109-155	1
516008	1109-197A	3	584323	1109-185	1
5250612	1130-4	1	584325	1109-184	1
528841-178	1109-35				

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per. Assy	Part No.	Fig. and Index No.	Qty per. Assy
584327	1109-143	1	61-30198-059	1117-17	1
594497	1109-43	1	61-30198-060	1117-17	1
6A47-003	1135-1	1	61-30198-061	1117-17	1
6B47-000	1135-1	1	61-30198-062	1117-17	1
60023	1124-4	1	61-30198-063	1117-17	1
60023	1125-10	1	61-30198-064	1117-17	1
60024	1124-4	1	61-30198-065	1117-17	1
61-30194-003	1117-12	1	61-30198-066	1117-17	1
61-30196-001	1117-13	1	61-30198-067	1117-17	1
61-30196-003	1117-13	1	61-30198-068	1117-17	1
61-30196-005	1117-13	1	61-30198-069	1117-17	1
61-30198-005	1117-17	1	61-30198-070	1117-17	1
61-30198-006	1117-17	1	61-30198-071	1117-17	1
61-30198-007	1117-17	1	61-30198-072	1117-17	1
61-30198-009	1117-17	1	61-30198-073	1117-17	1
61-30198-011	1117-17	1	61-30198-074	1117-17	1
61-30198-012	1117-17	1	61-30198-075	1117-17	1
61-30198-015	1117-17	1	61-30198-076	1117-17	1
61-30198-016	1117-17	1	61-30198-077	1117-17	1
61-30198-019	1117-17	1	61-30198-079	1117-17	1
61-30198-021	1117-17	1	61-30198-080	1117-17	1
61-30198-022	1117-17	1	61-30198-081	1117-17	1
61-30198-024	1117-17	1	61-30198-082	1117-17	1
61-30198-025	1117-17	1	61-30198-083	1117-17	1
61-30198-026	1117-17	1	61-30198-084	1117-17	1
61-30198-027	1117-17	1	61-30198-085	1117-17	1
61-30198-028	1117-17	1	61-30198-088	1117-17	1
61-30198-029	1117-17	1	61-30198-093	1117-17	1
61-30198-030	1117-17	1	61-30198-097	1117-17	1
61-30198-031	1117-17	1	61-30198-099	1117-17	1
61-30198-032	1117-17	1	61-30198-100	1117-17	1
61-30198-033	1117-17	1	61-30198-105	1117-17	1
61-30198-034	1117-17	1	61-30198-107	1117-17	1
61-30198-035	1117-17	1	61-30198-110	1117-17	1
61-30198-036	1117-17	1	61-30198-111	1117-17	1
61-30198-037	1117-17	1	61-30198-112	1117-17	1
61-30198-038	1117-17	1	61-30198-113	1117-17	1
61-30198-039	1117-17	1	61-30198-114	1117-17	1
61-30198-040	1117-17	1	61-30198-115	1117-17	1
61-30198-041	1117-17	1	61-30198-116	1117-17	1
61-30198-044	1117-17	1	61-30198-117	1117-17	1
61-30198-045	1117-17	1	61-30198-118	1117-17	1
61-30198-047	1117-17	1	61-30198-119	1117-17	1
61-30198-054	1117-17	1	61-30198-120	1117-17	1
61-30198-056	1117-17	1	61-30198-121	1117-17	1
61-30198-057	1117-17	1	61-30198-122	1117-17	1
61-30198-058	1117-17	1	61-30198-123	1117-17	1

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per. Assy
61-30198-124	1117-17	1
61-30198-125	1117-17	1
61-30198-126	1117-17	1
61-30198-127	1117-17	1
61-30198-128	1117-17	1
61-30198-129	1117-17	1
61-30198-130	1117-17	1
61-30198-131	1117-17	1
61-30198-132	1117-17	1
61-30198-133	1117-17	1
61-30198-134	1117-17	1
61-30198-135	1117-17	1
61-30198-136	1117-17	1
61-30198-137	1117-17	1
61-30198-138	1117-17	1
61-30198-140	1117-17	1
61-30198-141	1117-17	1
61-30198-142	1117-17	1
61-30198-143	1117-17	1
61-30198-144	1117-17	1
61-30198-145	1117-17	1
61-30198-146	1117-17	1
61-30198-147	1117-17	1
61-30198-148	1117-17	1
61-30198-149	1117-17	1
61-30198-150	1117-17	1
61-30198-151	1117-17	1
61-30198-152	1117-17	1
61-30198-153	1117-17	1
61-30198-154	1117-17	1
61-30198-155	1117-17	1
61-30198-156	1117-17	1
61-30198-157	1117-17	1
61-30198-158	1117-17	1
61-30198-159	1117-17	1
61-30198-160	1117-17	1
61-30198-161	1117-17	1
61-30198-162	1117-17	1
61-30198-163	1117-17	1
61-30198-164	1117-17	1
61-30198-165	1117-17	1
61-30198-166	1117-17	1
61-30198-167	1117-17	1
61-30198-168	1117-17	1
61-30198-169	1117-17	1
61-30198-170	1117-17	1
61-30198-171	1117-17	1
61-30198-172	1117-17	1
61-30198-173	1117-17	1

Part No.	Fig. and Index No.	Qty per. Assy
61-30198-174	1117-17	1
61-30198-175	1117-17	1
61-30198-176	1117-17	1
61-30198-177	1117-17	1
61-30198-178	1117-17	1
61-30198-179	1117-17	1
61-30198-181	1117-17	1
61-30198-185	1117-17	1
61-30198-191	1117-17	1
61-30198-193	1117-17	1
61-30198-195	1117-17	1
61-30198-200	1117-17	1
61-30198-201	1117-17	1
61-30198-202	1117-17	1
61-30198-204	1117-17	1
61-30198-205	1117-17	1
61-30198-206	1117-17	1
61-30198-207	1117-17	1
61-30198-208	1117-17	1
61-30198-209	1117-17	1
61-30198-211	1117-17	1
61-30198-213	1117-17	1
61-30198-214	1117-17	1
61-30198-215	1117-17	1
61-30198-216	1117-17	1
61-30198-217	1117-17	1
61-30198-218	1117-17	1
61-30198-222	1117-17	1
61-30198-223	1117-17	1
61-30198-224	1117-17	1
61-30198-226	1117-17	1
61-30198-227	1117-17	1
61-30198-228	1117-17	1
61-30198-229	1117-17	1
61-30198-230	1117-17	1
61-30198-231	1117-17	1
61-30198-232	1117-17	1
61-30198-233	1117-17	1
61-30198-234	1117-17	1
61-30198-235	1117-17	1
61-30198-236	1117-17	1
61-30198-237	1117-17	1
61-30198-238	1117-17	1
61-30198-239	1117-17	1
61-30198-240	1117-17	1
61-30198-241	1117-17	1
61-30198-242	1117-17	1
61-30198-243	1117-17	1

Part No.	Fig. and Index No.	Qty per. Assy
61-30198-244	1117-17	1
61-30198-245	1117-17	1
61-30198-246	1117-17	1
61-30198-247	1117-17	1
61-30198-248	1117-17	1
61-30198-249	1117-17	1
61-30198-250	1117-17	1
61-30198-251	1117-17	1
61-30198-252	1117-17	1
61-30198-253	1117-17	1
61-30198-254	1117-17	1
61-30198-255	1117-17	1
61-30198-256	1117-17	1
61-30198-257	1117-17	1
61-30198-258	1117-17	1
61-30198-259	1117-17	1
61-30198-260	1117-17	1
61-30198-261	1117-17	1
61-30198-262	1117-17	1
61-30198-263	1117-17	1
61-30198-264	1117-17	1
61-30198-267	1117-17	1
61-30198-268	1117-17	1
61-30198-269	1117-17	1
61-30198-270	1117-17	1
61-30198-271	1117-17	1
61-30198-273	1117-17	1
61-30198-274	1117-17	1
61-30198-275	1117-17	1
61-30198-276	1117-17	1
61-30198-277	1117-17	1
61-30198-278	1117-17	1
61-30198-280	1117-17	1
61-30198-281	1117-17	1
61-30198-282	1117-17	1
61-30198-283	1117-17	1
61-30198-284	1117-17	1
61-30198-285	1117-17	1
61-30198-286	1117-17	1
61-30198-289	1117-17	1
61-30198-290	1117-17	1
61-30198-291	1117-17	1
61-30198-292	1117-17	1
61-30198-293	1117-17	1
61-30198-294	1117-17	1

Part No.	Fig. and Index No.	Qty per. Assy
61-30198-295	1117-17	1
61-30198-296	1117-17	1
61-30198-297	1117-17	1
61-30198-298	1117-17	1
61-30198-299	1117-17	1
61-30198-300	1117-17	1
61-30198-301	1117-17	1
61-30198-302	1117-17	1
61-30198-303	1117-17	1
61-30198-304	1117-17	1
61-30198-305	1117-17	1
61-30198-306	1117-17	1
61-30198-307	1117-17	1
61-30198-308	1117-17	1
61-30198-309	1117-17	1
61-30198-310	1117-17	1
61-30198-311	1117-17	1
61-30198-312	1117-17	1
61-30198-313	1117-17	1
61-30198-314	1117-17	1
61-30198-315	1117-17	1
61-30198-316	1117-17	1
61-30198-317	1117-17	1
61-30198-318	1117-17	1
61-30198-319	1117-17	1
61-30198-320	1117-17	1
61-30198-321	1117-17	1
61-30198-322	1117-17	1
61-30198-323	1117-17	1
61-30198-324	1117-17	1
61-30198-325	1117-17	1
61-30198-326	1117-17	1
61-30198-327	1117-17	1
61-30198-328	1117-17	1
61-30198-329	1117-17	1
61-30198-330	1117-17	1
61-30198-331	1117-17	1
61-30198-332	1117-17	1
61-30198-333	1117-17	1
61-30198-335	1117-17	1
61-30198-336	1117-17	1
61-30198-337	1117-17	1
61-30198-338	1117-17	1
61-30198-339	1117-17	1
61-30198-340	1117-17	1

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
61-30198-341	1117-17	1	61-30198-430	1117-17	1
61-30198-342	1117-17	1	61-30198-431	1117-17	1
61-30198-343	1117-17	1	61-30198-432	1117-17	1
61-30198-344	1117-17	1	61-30198-433	1117-17	1
61-30198-345	1117-17	1	61-30198-434	1117-17	1
61-30198-346	1117-17	1	61-30198-435	1117-17	1
61-30198-347	1117-17	1	61-30198-436	1117-17	1
61-30198-348	1117-17	1	61-30198-437	1117-17	1
61-30198-349	1117-17	1	61-30198-438	1117-17	1
61-30198-350	1117-17	1	61-30198-439	1117-17	1
61-30198-351	1117-17	1	61-30198-440	1117-17	1
61-30198-352	1117-17	1	61-30198-441	1117-17	1
61-30198-353	1117-17	1	61-30198-446	1117-17	1
61-30198-354	1117-17	1	61-30198-447	1117-17	1
61-30198-355	1117-17	1	61-30198-448	1117-17	1
61-30198-356	1117-17	1	61-30198-449	1117-17	1
61-30198-357	1117-17	1	61-30198-450	1117-17	1
61-30198-358	1117-17	1	61-30198-453	1117-17	1
61-30198-359	1117-17	1	61-30200-001	1117-14	1
61-30198-360	1117-17	1	61-30200-003	1117-14	1
61-30198-361	1117-17	1	61-30200-005	1117-14	1
61-30198-364	1117-17	1	61-30200-007	1117-14	1
61-30198-366	1117-17	1	61-30202-001	1117-15	1
61-30198-368	1117-17	1	61-30204-001	1117-16	1
61-30198-369	1117-17	1	61-30204-002	1117-16	1
61-30198-370	1117-17	1	61-30204-2	1117-16	1
61-30198-371	1117-17	1	61-30204-3	1117-16	1
61-30198-372	1117-17	1	61-30404-001	1117-12	1
61-30198-373	1117-17	1	61-30404-002	1117-12	1
61-30198-374	1117-17	1	61-30404-1	1117-12	1
61-30198-375	1117-17	1	61-30404-2	1117-12	1
61-30198-380	1117-17	1	61-30404-3	1117-12	1
61-30198-381	1117-17	1	61-30404-011	1117-12	1
61-30198-383	1117-17	1	613780	1109-104	1
61-30198-395	1117-17	1	613781	1109-42A	1
61-30198-402	1117-17	1	624A65HD	1114-16	12
61-30198-403	1117-17	1	63-97273-2	1128-20	2
61-30198-404	1117-17	1	639239	1109-2	1
61-30198-409	1117-17	1		1109-87	1
61-30198-412	1117-17	1	639244	1109-30	1
61-30198-413	1117-17	1	639245	1109-47	1
61-30198-417	1117-17	1	639247	1109-48	1
61-30198-424	1117-17	1	639256	1109-66	1
61-30198-425	1117-17	1	639258	1109-73	1
61-30198-426	1117-17	1	639260	1109-70	1
61-30198-427	1117-17	1	639261	1109-84	1
61-30198-428	1117-17	1	639263	1109-90	1
61-30198-429	1117-17	1	639264	1109-87	1
			639266	1109-119	1

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
639267	1109-117	1	65-21950-7	1104	
639268	1109-116	1	65-21950-8	1104	
639269	1109-112	1	65-22249-1	1105	
639273	1109-204	1	65-22249-2	1105	
639274	1109-205	1	65-22249-3	1105	
639275	1109-203	1	65-22429-15	1122-1	1
647554	1109-28	1	65-22429-19	1122-1	1
647555	1109-29	1	65-22429-22	1122-1	1
647557	1109-20C	2	65-22441-21	1111-1	1
647557	1109-20D	1	65-22441-24	1111-1	1
647557	1109-20E	1	65-22441-28	1111-1	1
647557	1109-31	7	65-22441-31	1111-1	1
647557	1109-32	1	65-22441-34	1111-1	1
647557	1109-33	2	65-22441-39	1111-1	1
647557	1109-34	1	65-22454-3	1106	
647557	1109-49	2	65-22454-7	1106	
647560	1109-179	1	65-22454-8	1106	
647562	1109-178	1	65-22454-9	1106	
647567	1109-38	1	65-22454-13	1106	
647569	1109-54	1	65-22454-14	1106	
65-19592-23	1101	RF	65-22454-15	1106	
65-19592-24	1101	RF	65-22454-16	1106	
65-19592-25	1101-15	1	65-22454-21	1106	
65-19592-3	1101	RF	65-22454-20	1106	
65-19684-1	1102	RF	65-22454-24	1106	
65-19684-17	1102-2	1	65-22454-25	1106	
65-19684-2	1102-1	1	65-22486-114	1109-37	1
65-19684-3	1102-6	1	65-22486-135	1109-82E	1
65-20996-1	1103	RF	65-22486-137	1109-18	2
65-20996-9	1103	RF	65-22486-137	1109-29D	1
65-20996-10	1103	RF	65-22486-141	1109-29	1
65-20996-11	1103	RF	65-22486-142	1109-28	1
65-20996-17	1103	RF	65-22486-148	1109-60	1
65-20996-19	1103	RF	65-22486-16	1109-97	1
65-20996-21	1103	RF	65-22486-169	1109-142	1
65-20996-22	1103	RF	65-22486-173	1109-65	1
65-20996-23	1103	RF	65-22486-195	1109-8	7
65-20996-24	1113	RF	65-22486-206	1109-19	3
65-20996-25	1113	RF	65-22486-206	1109-20	1
65-21931-18	1114A	RF	65-22486-206	1109-20A	1
65-21931-21	1114A	RF	65-22486-206	1109-23	3
65-21931-22	1114A	RF	65-22486-206	1109-23A	1
65-21931-23	1114A	RF	65-22486-230	1109-62	1
65-21931-24	1114A	RF	65-22486-258	1109-2A	3
65-21950-2	1104			1109-3	7
65-21950-3	1104		65-22486-260	1109-6A	1
65-21950-5	1104		65-22486-30	1109-2A	7
65-21950-6	1104				

Part No.	Fig. and Index No.	Qty per. Assy
65-22486-30	1109-3	10
65-22486-32	1109-6A	1
65-22486-33	1109-98	1
65-22486-33	1109-102	2
65-22486-33	1109-104A	1
65-22486-35	1109-124	1
65-22486-50	1109-74	1
65-22486-8	1109-78	1
65-22486-86	1109-123	1
65-22486-87	1109-47	1
65-22486-8	1109-32	1
65-22486-88	1109-48	1
65-22486-93	1109-25	1
65-22486-95	1109-59	2
65-25736-2	1107	1
65-25736-3	1107	1
65-27808-14	1108-1	1
65-27808-800	1108-1	1
65-45798-1	1109	
65-45798-2	1109	
65-45798-3	1109	
65-45798-4	1109	
65-45798-5	1109	
65-45798-6	1109	
65-45798-7	1109	
65-45798-8	1109	
65-45798-9	1109	
65-45798-10	1109	
65-45798-800	1109-137	
65-45798-801	1109-138	
65-45798-802	1109-139	
65-45799-1	1109	
65-45799-10	1109-208	1
65-45799-100	1109-90	1
65-45799-101	1109-151	1
65-45799-102	1109-179	1
65-45799-103	1109-15	1
65-45799-105	1109-117	1
65-45799-109	1109-104	1
65-45799-11	1109-114	1
65-45799-11	1109-114C	1
65-45799-110	1109-42A	1
65-45799-117	1109	
65-45799-118	1109-64	1
65-45799-12	1109-114B	2
65-45799-12	1109-114E	1
65-45799-120	1109	

Part No.	Fig. and Index No.	Qty per. Assy
65-45799-121	1109-3A	1
65-45799-122	1109-155	1
65-45799-123	1109	
65-45799-124	1109-197	1
65-45799-125	1109-198	1
65-45799-126	1109-4B	1
65-45799-127	1109	
65-45799-128	1109-14A	1
65-45799-129	1109-23B	1
65-45799-13	1109-114A	2
65-45799-13	1109-114D	1
65-45799-130	1109-24A	1
65-45799-131	1109-33A	1
65-45799-132	1109-64	1
65-45799-133	1109-155	1
65-45799-134	1109	
65-45799-135	1109-114B	1
65-45799-136	1109-114E	1
65-45799-137	1109-114	1
65-45799-138	1109-114C	1
65-45799-140	1109-7	1
65-45799-141	1109-23B	1
65-45799-142	1109-78A	1
65-45799-143	1109-155	1
65-45799-144	1109-64	1
65-45799-145	1109-14A	1
65-45799-147	1109	
65-45799-148	1109	
65-45799-149	1109-3A	1
65-45799-15	1109-104	1
65-45799-150	1109	RF
65-45799-151	1109	RF
65-45799-152	1109-8A	4
65-45799-153	1109-2	1
65-45799-154	1109-14	2
65-45799-155	1109-4	1
65-45799-156	1109-17	1
65-45799-157	1109-7	1
65-45799-158	1109-8C	2
65-45799-159	1109-10	1
65-45799-16	1109-17	1
65-45799-160	1109-15	1
65-45799-161	1109-3A	1
65-45799-162	1109-14A	1
65-45799-163	1109-8B	1
65-45799-164	1109-4C	1
65-45799-165	1109	

Part No.	Fig. and Index No.	Qty per. Assy
65-45799-166	1109	
65-45799-167	1109-185	1
65-45799-168	1109-204	1
65-45799-169	1109-205	1
65-45799-17	1109-120	1
65-45799-170	1109	RF
65-45799-171	1109	RF
65-45799-172	1109-3A	1
65-45799-173	1109-68H	1
65-45799-174	1109-64	1
65-45799-19	1109-117	1
65-45799-20	1109-184	1
65-45799-21	1109-85	1
65-45799-22	1109-61	1
65-45799-23	1109-7	1
65-45799-24	1109-66	1
65-45799-25	1109-26	1
65-45799-26	1109-183	1
65-45799-27	1109-83	5
65-45799-27	1109-86	1
65-45799-28	1109-67	1
65-45799-29	1109-4	1
65-45799-3	1109-95	1
65-45799-30	1109-42	3
65-45799-32	1109-38	1
65-45799-33	1109-43	1
65-45799-34	1109-68	1
65-45799-35	1109-35	2
65-45799-36	1109-69	1
65-45799-37	1109-77	1
65-45799-38	1109-143	1
65-45799-39	1109-63	1
65-45799-4	1109-207	1
65-45799-40	1109-75	1
65-45799-41	1109-74	2
65-45799-43	1109-9	2
65-45799-45	1109-185	1
65-45799-46	1109-36	1
65-45799-47	1109-89	1
65-45799-47	1109-94	1
65-45799-48	1109-16	1
65-45799-49	1109-118	1
65-45799-5	1109-80	1
65-45799-50	1109-76	1
65-45799-51	1109-20B	1
	1109-20D	1
65-45799-52	1109-64	1

Part No.	Fig. and Index No.	Qty per. Assy
65-45799-55	1109-24	1
65-45799-57	1109-155	1
65-45799-58	1109-34A	1
65-45799-59	1109-15	1
65-47599-6	1109-71	1
65-45799-64	1109-82	1
65-45799-65	1109-20	1
65-45799-65	1109-20C	2
65-45799-65	1109-20D	2
	1109-20E	1
65-45799-65	1109-49	2
65-45799-65	1109-31	7
65-45799-65	1109-32	1
65-45799-65	1109-33	2
65-45799-65	1109-34	1
65-45799-7	1109-80A	1
65-45799-71	1109-116	1
65-45799-72	1109-84	1
65-45799-73	1109-30	1
65-45799-74	1109-70	1
65-45799-75	1109-197A	3
65-45799-76	1109-54	1
65-45799-77	1109-119	1
65-45799-8	1109-71A	1
65-45799-80	1109-203	1
65-45799-800	1109-113	6
65-45799-800	1109-121	1
65-45799-800	1109-122	5
65-45799-800	1109-123A	1
65-45799-800	1109-124A	1
65-45799-801	1109-199	2
65-45799-801	1109-206	1
65-45799-802	1109-163	1
65-45799-805	1109-10	1
65-45799-81	1109-2	1
	1109-87	1
65-45799-82	1109-73	1
65-45799-875	1109-204	1
65-45799-876	1109-205	1
65-45799-9	1109-81	1
65-45799-90	1109-112	1
65-45799-92	1109-198A	1
65-45799-93	1109-178	1
65-45799-94	1109-54	1
65-45799-95	1109-28	1
65-45799-96	1109-29	1
65-45799-97	1109-47	1

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per. Assy
65-45799-98	1109-48	1
65-45799-99	1109-87	1
65-46505-1	1122-2	1
65-46505-3	1122-2	1
65-46851-1	1111	RF
65-46851-2	1111	RF
65-46851-3	1111	RF
65-46851-4	1111	RF
65-46851-5	1111	RF
65-46852-1	1112	RF
65-46852-2	1112	RF
65-46858-1135	1138-36	1
65-46858-1136	1138-36A	1
65-46858-1139	1138-37	1
65-46858-1140	1138-37A	1
65-46858-131	1138	RF
65-46858-132	1138	RF
65-46858-135	1138-36	1
65-46858-136	1138-36A	1
65-46858-139	1138-37	1
65-46858-140	1138-37A	1
65-46858-187	1138	RF
65-46858-188	1138	RF
65-46858-273	1138	RF
65-46858-274	1138	RF
65-46869-4	1126-1	1
65-46869-800	1126-1	1
65-48205-3	1109-11	1
65-50545-1	1110-5	1
65-50545-3	1110-5	1
65-50546-1	1110-6	1
65-50571-1	1114	RF
65-50571-2	1114	RF
65-50571-4	1114	RF
65-50571-6	1114	RF
65-50571-7	1114	RF
65-50889-1	1113-28	1
65-51517-1	1110-1	1
65-51517-3	1110-1	1
65-51534-1	1110	1
65-51534-10	1110-6A	1
65-51534-11	1110-6A	1
65-51534-12	1110-20	1
65-52971-1	1117	RF
65-52971-5001	1117	RF
65-52971-5002	1117	RF
65-52971-5003	1117	RF
65-52971-5004	1117	RF
65-52971-5005	1117	RF

Part No.	Fig. and Index No.	Qty per. Assy
65-52971-5006	1117	RF
65-52971-5007	1117	RF
65-52971-5008	1117	RF
65-52971-5009	1117	RF
65-52971-5010	1117	RF
65-52971-5011	1117	RF
65-52971-5012	1117	RF
65-52971-5013	1117	RF
65-52971-5014	1117	RF
65-52971-5015	1117	RF
65-52971-5016	1117	RF
65-52971-5017	1117	RF
65-52971-5018	1117	RF
65-52971-5019	1117	RF
65-52971-5020	1117	RF
65-52971-5021	1117	RF
65-52971-5022	1117	RF
65-52971-5023	1117	RF
65-52971-5024	1117	RF
65-52971-5025	1117	RF
65-53863-1	1112-35	1
65-53863-22	1112-35	1
65-53863-24	1112-59	1
65-53863-28	1112-35	1
65-53863-30	1112-59	1
65-53863-32	1112-35	1
65-53863-802	1112-59	1
65-53864-800	1111-55	1
65-53865-1	1111-17	1
65-53865-802	1111-23	1
65-53865-803	1111-23	1
65-53866-16	1111-33	1
65-53866-800	1111-33	1
65-53867-1	1113-31	1
65-53867-2	1113-32	1
65-53867-4	1113-31	1
65-54476-1	1118	RF
65-54476-2	1118	RF
65-54476-3	1118	RF
65-54476-4	1118	RF
65-55475-1	1119	RF
65-55475-11	1119-1	1
65-55475-13	1119-42	1
65-55475-14	1119-43	1
65-55475-16	1119-44	1
65-55475-21	1119	RF
65-55475-22	1119-26	1
65-55475-23	1119-32	1

Part No.	Fig. and Index No.	Qty. per Assy.
65-55475-24	1119-24	1
65-55475-25	1119-8	1
65-55475-26	1119	RF
65-55475-27	1119-1	1
65-55475-28	1119	RF
65-55475-29	1119-25	1
65-55475-4	1119-14	1
65-55475-5	1119-18	1
65-55475-8	1119-25	1
65-55475-802	1119-46	1
65-55475-803	1119-47	1
65-55749-4	1120-1	2
65-55749-5	1120-2	2
65-55749-6	1120-1	2
65-55835-13	1138	RF
65-55835-14	1138	RF
65-55835-5	1138	RF
65-55835-6	1138	RF
65-55860-1	1122-28	1
65-55860-3	1122-28	1
65-56646-1	1122	1
65-56646-2	1122	1
65-56646-3	1122	1
65-57814-1	1140-1	1
65-57814-2	1140-1	1
65-57814-5	1140-1	1
65-57814-6	1140-1	1
65-57815-1	1140	1
65-57815-2	1140	1
65-58123-1	1125-7	1
65-58123-10	1125-7	1
65-58123-13	1125-8	1
65-58123-2	1125-8	1
65-58123-7	1125-9	1
65-58126-10	1125-35	1
65-58126-11	1125-32	1
65-58126-12	1125-41	1
65-58126-17	1125-54	1
65-58126-18	1125-54	1
65-58126-19	1125-61	1
65-58126-20	1125	RF
65-58126-21	1125-33A	1
65-58126-23	1125-15A	1
65-58126-24	1125-92	1
65-58126-25	1125-92	1
65-58126-26	1125-93	1
	1103-20	1
65-58126-27	1125-93	1
	1103-20	1

Part No.	Fig. and Index No.	Qty. per Assy.
65-58126-6	1125-34	1
65-58126-800	1125-33	1
65-58126-801	1125-32	1
65-58135-13	1126-12	1
65-58135-14	1126-24	1
65-58135-15	1126-37	1
65-58135-16	1126-38	1
65-58135-19	1126-58	1
65-58135-25	1126-58	1
65-58135-27	1126-12	1
65-58135-28	1126-10	1
65-58135-29	1126-10	1
65-58135-30	1126	RF
65-58135-31	1126	RF
65-58135-32	1126-43	1
65-58135-33	1126-44	1
65-58135-34	1126-45	1
65-58135-35	1126-46	1
65-58135-36	1126-36E	1
65-58135-37	1126	RF
65-58135-38	1126-50	1
65-58135-39	1126-51	1
65-58135-40	1126-51A	1
65-58135-41	1126-51B	1
65-58135-42	1126-51C	1
65-58135-43	1126-51D	1
65-58135-44	1126-51E	1
65-58135-45	1126-51F	1
65-58135-46	1126	RF
65-58135-47	1126	RF
65-58135-48	1126	RF
65-58135-4	1126	RF
65-58135-5	1126-9	1
65-58135-6	1126-10	1
65-58135-8	1126-11	1
65-58145-10	1127	RF
65-58145-11	1127-2	1
65-58145-12	1127	RF
65-58145-13	1127-2	1
65-58145-15	1127	RF
65-58145-3	1127-1	1
65-58145-5	1127-1	1
65-58145-7	1127-1	1
65-58145-8	1127-35	1
65-58234-10	1128	RF
65-58234-11	1128	RF
65-58234-15	1128	RF
65-58234-16	1128	RF
65-58234-17	1128	RF

POWERPLANT BUILDUP

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty. per Assy.
65-58234-18	1128	RF
65-58234-19	1128	
65-58234-5003	1128	RF
65-58234-5006	1128	RF
65-58234-5007	1128	RF
65-60529-673	1121-1	1
65-60529-674	1121-1	1
65-60529-767	1121-1	1
65-60529-768	1121-1	1
65-60529-771	1121-1	1
65-60529-772	1121-1	1
65-60529-787	1121-1	1
65-60529-788	1121-1	1
65-60529-1499	1121-1	1
65-60529-1500	1121-1	1
65-61732-1	1137-2	1
65-63375-1	1137-2	1
65-64491-1	1117-22	1
65-64491-2	1117-22	1
65-65358-1	1117-2	1
65-65358-2	1117-4	1
65-65919-4	1108	RF
65-65919-5	1108-1	1
65-6758-1	1124-10	1
65-68621-1	1121-1	1
65-68621-27	1121-1	1
65-69659-1	1142A	RF
65-69659-10	1142A	RF
65-69659-11	1142A	RF
65-69659-12	1142A	RF
65-69659-13	1142A	RF
65-69659-14	1142A	RF
65-69659-15	1142A	RF
65-69659-16	1142A	RF
65-69659-2	1142A	
65-69659-4	1142A	
65-69659-5	1142A	
65-69659-6	1142A	
65-69659-7	1142A	
65-69659-8	1142A	RF
65-69659-9	1142A	RF
65-70190-1	1138-3	1
65-70191-1	1138-1	1
65-70191-34	1138-1	1
65-70197-4	1138-4	1
65-70950-1	1138	
65-70950-11	1138	
65-70950-12	1138	
65-70950-15	1138	

Part No.	Fig. and Index No.	Qty. per Assy.
65-70950-16	1138	
65-70950-2	1138	
65-70950-21	1138	
65-70950-22	1138	
65-70950-23	1138	
65-70950-24	1138	
65-70950-3	1138	
65-70950-4	1138	
65-70950-5	1138	
65-70950-6	1138	
65-70950-7	1138	
65-70950-8	1138	
65-77451-1	1125	
65-77451-2	1125	
65-77451-3	1125	
65-77451-4	1125	
65-77452-1	1125-78	1
65-77452-10	1125-72	1
65-77452-16	1125-72	1
65-77452-13	1125-78	1
65-82788-1	1138-3	1
65-82788-6	1138-3	1
65-85369-1	1143-2	1
65-85369-11	1143-2	1
65-85369-12	1143-2	1
65-85369-13	1143-2	1
65-85369-14	1143-2	1
65-85369-15	1143-2	1
65-85369-16	1143-2	1
65-85369-2	1143-2	1
65-85369-21	1143-2	1
65-85369-22	1143-2	1
65-85369-3	1143-2	1
65-85369-4	1143-2	1
65-85378-1	1143-1	1
65-85378-101	1143-1	1
65-85378-102	1143-1	1
65-85378-2	1143-1	1
65-85378-311	1143-1	1
65-85378-312	1143-1	1
65-85378-315	1143-1	1
65-85378-316	1143-1	1
65-85378-359	1143-1	1
65-85378-360	1143-1	1
65-85378-383	1143-1	1
65-85378-384	1143-1	1
65-85378-387	1143-1	1

Part No.	Fig. and Index No.	Qty. per Assy	Part No.	Fig. and Index No.	Qty per. Assy
65-85378-388	1143-1	1	69-15863-1	1130	RF
65-85378-391	1143-1	1	69-15863-10	1130-7	1
65-85378-392	1143-1	1	69-15863-21	1130A-4	1
65-85378-395	1143-1	1	69-15863-5001	1130	RF
65-85378-396	1143-1	1	69-15863-5002	1130	RF
65-85378-397	1143-1	1	69-15863-5003	1130	RF
65-85378-398	1143-1	1	69-15863-8	1130-7	1
65-85379-1	1143		69-15863-9	1130-7	1
65-85379-2	1143		69-17969-3	1112-36	1
65-85379-3	1143	RF	69-17969-3	1112-63	1
65-85379-4	1143	RF	69-18491-4	1102-3	1
65-85379-5	1143	RF	69-18713-800	1102-13	1
65-85379-6	1143	RF	69-18713-800	1143-3	1
65-86025-10	1143-1	1	69-20607-6	1109-186	1
65-86025-9	1143-1	1		1109-82J	1
65C13014-1	1103-16	1		1109-97A	1
65C13014-3	1103-16	1	69-20823-1	1112-53	1
65C13050-1	1104-5	1	69-21612-1	1132	1
65C13050-2	1104-6	1	69-22313-12	1109-156	1
65C15618-2	1138-11	1		1109-68G	1
65C20874-1	1122-14	1	69-22313-13	1109-159	1
65C21843-1	1138-36	1		1109-82F	1
65C21843-2	1138-36A	1	69-24228-1	1109-196A	1
65C21843-3	1138-37	1		1109-82K	1
65C21843-4	1138-37A	1	69-25331-1	1109-145	1
66-12113-1	1130-22	1		1109-29	1
66-12113-1	1133-23	1	69-25331-2	1109-147	1
66-13297-2	1101-7	1		1109-28	1
66-13955-1	1115-2A	1	69-25331-5	1109-146	1
66-13955-857	1115-2A	1		1109-28	1
66-13955-857	1115-27	1	69-25331-8	1109-144	1
66-14330-1	1137-1	1		1109-29	1
66-14598-1	1111-28	1	69-26229-2	1111-18	1
66-15218-1	1134-1	1	69-26229-2	1112-37	1
66-15227-5	1105-21	1	69-26229-2	1112-60	1
66-15330-1	1107-18	2	69-26230-2	1111-56	1
66-16394-1	1132-2	1	69-26230-2	1112-54	1
66-16394-2	1132-3	1	69-26231-2	1112-2	2
66-16435-1	1132-1	2	69-28727-1	1133	RF
66-18526-1	1117-11	1	69-28727-3	1133-4	1
66-19214-1	1115-21	1	69-28727-5	1133-4	1
66-20269-1	1103-8	1	69-28727-5001	1133	RF
66-20269-2	1103-8	1	69-28727-5002	1133	RF
66-20269-800	1103-8	1	69-28727-8	1133	RF
66-8324-1	1124-14	1	69-28727-9	1133-4	1
66081	1125-15A	1	69-29863-2	1109-131	1
67-2905-001	1112-56	1	69-29863-2	1101-31	1
689460A	1114-3	1			

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
69-31608-1	1134-6	1	69-43468-1	1109-125	2
69-31653-1	1107-18	2	69-43487-1	1109-126	8
69-31661-1	1132-2	1	69-43481-2	1110-13	2
69-31661-2	1132-3	1	69-44477-1	1124-10	1
69-31696-1	1122-10	1	69-44633-3	1112-5	1
69-33197-2	1109-82A	1	69-44633-3	1112-20	1
69-36670-1	1134	RF	69-44634-1	1112-18	1
69-36670-2	1134	RF	69-44634-4	1112-18	1
69-35244-1	1105-8	1	69-44635-1	1112-1	1
69-36634-4	1109-209	1	69-45172-1	1136	
69-36671-1	1134-3	1	69-45172-2	1136-1	1
69-38973-1	1116-2	1	69-45172-3	1136	
69-39486-4	1116-13	1	69-45172-4	1136-1A	1
69-39487-1	1116-7	1	69-45174-1	1111-11	1
69-39494-1	1116-1	1	69-45175-1	1111-5	1
69-39494-1	1116-6	1	69-45402-1	1122-3	1
69-39494-2	1116-11	1	69-45402-10	1109-166A	1
69-39688-1	1109-160	1	69-45402-11	1109-26B	1
	1109-68F	1	69-45402-12	1109-35B	1
	1109-68G	1	69-45402-2	1109-102A	1
69-40075-2	1117-13L	1	69-45402-2	1109-200	1
69-40075-4	1109-13N	1	69-45402-7	1109-164	1
69-41777-1	1109-96	1	69-45402-9	1109-174A	1
69-41777-800	1109-96	1	69-45428-1	1109-99	1
69-41780-1	1109-88	1	69-45428-10	1109-99	1
69-42533-1	1113-17	1	69-45428-12	1109-99	1
69-42569-3	1113-9	1	69-45428-14	1109-110	1
69-42570-1	1113-1	1	69-45428-3	1109-190	1
69-42570-2	1113-1	1		1109-82L	1
69-42571-1	1113-5	1	69-45428-4	1109-110	1
69-42571-2	1113-5	1	69-45428-8	1109-194	1
69-42802-1	1111-42	1	69-45446-4	1109-105	1
69-42825-1	1113-19	2	69-45458-1	1115-22	1
69-42852-1	1109-27	1		1118-8	1
69-42852-800	1109-27	1	69-45492-1	1122-26	1
69-42961-1	1122-37	1	69-45492-2	1109-109	1
69-42961-6	1122-37	1	69-45492-3	1109-107	1
69-43284-5	1122-20		69-45804-2	1107-1	1
69-43284-6	1122-19	2	69-45804-3	1107-2	1
69-43284-7	1122-20D	1	69-45804-4	1109-172	1
69-43284-8	1122-20A	1	69-46653-1	1113-16	1
69-43285-1	1122-18	1	69-46654-1	1113-15	1
69-43416-1	1109-79	2	69-48205-3	1109-11	1
69-43417-1	1109-111	1	69-48235-1	1109-136	
69-43424-3	1112-70	1	69-48235-4	1109-155A	1
69-43424-4	1112-69	1	69-48904-1	1119-57	1
69-43424-5	1112-70	1	69-48918-6	1128-9	1
69-43424-6	1112-69	1	69-48918-7	1128-2	1
69-43437-1	1111-24	1			

OVERHAUL MANUAL

Part No.	Fig. and Index No.	Qty per Assy	Part No.	Fig. and Index No.	Qty per Assy
69-48918-8	1128-9	1	69-56508-6	1112-62	1
69-48936-1	1137	1	69-58497-1	1109-186	1
69-49832-1	1140-2	2		1109-82I	1
69-49854-1	1138-16	3		1109-97A	1
69-49854-3	1138-16	3	69-58497-6	1109-82H	1
69-50910-1	1141-4	1		1109-97A	1
69-50910-2	1141-5	1		1109-186	1
69-50910-3	1141-6	1	69-60439-1	1109-146	1
69-50910-4	1141-4	1		1109-28	1
69-50910-5	1141-6	1	69-60439-2	1109-144	1
69-50911-1	1141			1109-29	1
69-50911-2	1141		69-60758-1	1109-3B	1
69-50919-1	1141-4	1	69-60758-2	1109-3C	1
69-52217-1	1114-17	1	69-60758-3	1109-3C	1
69-52217-2	1122-17	1	69-62249-1	1109-108	1
69-52217-3	1122-4	1	69-62249-2	1116-12	1
69-52217-4	1134-2	1	69-62261-1	1113-31	1
69-52217-14	1134-2	1	69-62271-1	1109-209	1
69-52261-1	1109-216	1	69-62274-1	1113-28A	1
69-52261-3	1109-216A	1	69-62277-1	1116-7	1
69-52264-1	1117-18	1	69-64545-1	1126-31	1
69-52264-2	1117-19	1	69-64545-4	1126-31	1
69-52265-1	1117-23	1	69-64750-2	1111-56	1
69-52283-1	1116-12	1	69-64750-3	1111-18	1
69-52283-2	1128-4	1	69-65327-1	1133-22	1
69-52283-3	1109-108	1	69-65327-2	1130-21	1
69-52291-1	1109-219	1	69-65327-3	1130-21	1
69-52291-3	1109-219	1	69-65869-3	1109-190	1
69-52291-5	1109-219	1	69-66601-1	1109-117A	1
69-52295-1	1109-106	1	69-66635-1	1109-68C	1
69-52295-2	1109-91	1	69-66647-2	1110-16	1
69-52860-1	1117-3	1	69-66647-3	1110-11	1
69-53880-1	1109-167	1	69-67501-1	1133-22	1
69-53885-1	1109-167	1	69-67501-2	1130-21	1
69-53885-2	1109-167	1	69-67501-3	1130-21	1
69-56334-1	1139		69-71522-1	1109-35A	1
69-56508-1	1112-3	2	69-71535-1	1110-26	1
69-56508-2	1111-57	1	69-71542-1	1110-6B	1
69-56508-2	1112-51	1	69-71573-1	1114A-4	1
69-56508-3	1111-19	1	69-71727-1	114A-5	1
69-56508-3	1112-38	1	69-71727-2	114A-6	1
69-56508-3	1112-61	1	69-71727-8	1122-4A	1
69-56508-4	1112-4	2	69-71727-9	1114A-5	1
69-56508-5	1111-58	1	69-72053-1	1101-12	1
69-56508-5	1112-52	1	69-72053-2	1101-13	1
69-56508-6	1111-20	1	69-72053-3	1101-12	2
69-56508-6	1112-39	1	69-72053-4	1101-13	2

Part No.	Fig. and Index No.	Qty per. Assy	Part No.	Fig. and Index No.	Qty per. Assy
69-72056-1	1101-27	1	758093	1109-15	1
69-72056-2	1101-19	1	758097	1109-17	1
69-72056-3	1101-28	1	7581418	1123-1	1
69-72056-4	1101-19	1	773089	1105-5	1
69-72057-1	1101-23	1	762712	1105-5	1
69-72085-1	1109-68B	1	773083	1105-5	1
69-72085-2	1109-68A	1	8C150	1124-17	1
69-72085-3	1109-27C	1	8G654	1125-67	1
69-72688-1	1118-1A	1	8TJ50GBZ	1105-1	1
69-76789-1	1101-60	1	8TJ50GCA	1105-1	1
69-76789-4	1101-61	1	8TJ85GAZ	1105-1A	1
69-77002-1	1138-34A	1	8TJ85GHA	1105-1A	1
69-77057-1	1101-23	1	8023812	1130-4	1
69-78715-1	1101-62	1	856411-1	1112-65	1
69B00412-6	1101-16	AR	894093	1128-1	1
69B63073-501	1101-17	AR	894504	1128	1
693206	1114A-1C	2	898172-1-1	1103-2	1
693601	1114A-1H	1	898172-3-1	1103-2	1
693608	1114A-1E	1	899929	1128-1	1
694282	1114-22	1			
698464	1109-8E	1	9011520100-1	1130-19	1
698601	1109-8D	1	9011520100-1	1133-20	1
699647	1114-2	1	908D049-1	1114A-3G	1
699771	1114-23	1	918B105-2	1114A-3D	4
			938D844-3	1122-14	1
7026	1125-64	1	941D339-1	1117-1	1
709129	1109-24A	1	976J498-2	1114-1	1
709130	1109-23B	1	976J597-1	1114-1	1
709139	1109-33A	1	976J598-1	1114-1	1
709140	1109-54	1	979078-1-1	1103-2	1
706183	1114A-23	1	979078-3-1	1103-2	1
716038	1109-78A	1			
716039	1109-7	1			
730019	1114A-22	1			
7500203	1123-1	1			
7513182	1123-1A	1			
755765	1109-2A	3			
	1109-3	7			
755771	1109-6A	1			
755775	1109-1	1			
755782	1109-6	1			
755788	1109-4B	1			
	1109-6B	1			
758083	1109-2	1			
758085	1109-4	1			
758086	1109-7	1			
758089	1109-10	1			
758090	1109-14	2			

OVERHAUL MANUAL

QEC CONFIGURATION INDEX

INTRODUCTION

1. This section provides the information necessary to determine the various power plant configurations applicable to individual airlines. The data is included in four separate tables.
 - A. Table A -- Airline Designator Codes
 - (1) This table lists the airline name and related two-letter code. The codes are used to locate the appropriate Power Pack Installation in Table D.
 - B. Table B -- Engine Variables
 - (1) This table lists the engine designator numbers (JT8D BG dash numbers), the related engine variable installations, and the engine thrust rating. This information is used to determine which configurations are peculiar to specified engine components.
 - C. Table C -- Common Installations
 - (1) The original intent of this table was to list installations which were used on all 737 power plant buildups. However, with diversification of standard options, some installations are no longer common to all 737 buildups and are designated by *[1].
 - D. Installation Tables
 - (1) These tables list the variable Boeing installations applicable to each airline and the related Illustrated Parts List figure number. The tables are identified by airline code. The codes are listed in alphabetical order and located at the bottom and center of each page following Table C.

AIRLINE	CODE
Aer Lingus	IN
Aerolineas Argentinas	AR
Aerotron Aircraft Radio, Inc.	EO
Air Algerie (Compagnie Generale de Transports Aeriens)	AH
Air Belgium	EB
Air California, Inc.	AP
Air Executive Norway-Busy Bee A/S	BB
Air Europe, Ltd. (Intasun)	WU
Air Florida, Inc.	FF
Air France (Compagnie Nationale Air France)	AF
Air Gabon, S.A.	GN
Air Guinee	GI
Air Madagascar (Societe Nationale Malagache de Transports Aeriens)	MD
Air Malta	YB
Air Nauru	ON
Air New Zealand Limited	NZ
Air Pacific Limited	PC
Air Tanzania	TJ
Air Zaire	QC
Air Zimbabwe	RH
Alaska Airlines, Inc.	AS
Alaska International Airlines (Markair Inc.)	ALK
All Nippon Airways Co., Ltd.	NH
Aloha Airlines Inc.	TS
Alyemda Democratic Yemen Airlines (South Yemen)	DY
Angola Airlines (Linhas Aeras de Angola)	DT
Ansett Airlines of Australia	AN
Aramco Services Company	MR
Arkia (Israel Inland Airlines Limited)	ARK
Aviogenex Air Transport	JJ
Bahamas Air	UP
Bavaria Fluggesellschaft MBH	BL
Braathens S.A.F.E. Airtransport A/S	BU
Britannia Airways Ltd.	ZD
British Airways, Ltd.	BA
British Airtours, Ltd.	BAT
Cameroon Airlines	VM
China Airlines, Ltd.	CI
Civil Aviation Administration of China (CAAC)	CA
CP Air (Canadian Pacific)	CP
Cruzeiro (Servicos Aereos Cruzeiro do Sul, S.A.)	Deleted
Delta Airlines, Inc.	DL
D.E.T.A. (Direccao de Exploracao dos Transportes Aeroes) - Linhas Aereas de Mozambique	TM
Deutsche Lufthansa AG	LH

TABLE A

Airline Designator Codes
(Sheet 1)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE	CODE
Eastern Provincial Airways Ltd. (EPA)	PV
Egypt Air	MS
EL AL Israel Airlines, Ltd.	LY
Eldorado Aviation, Ltd.	EL
Essex Wire Corp.	EZ
Euralair	EK
Far Eastern Air Transport Corp.	EF
Federal Express	FE
Frontier Airlines, Inc.	FL
Gulf Aviation Company Ltd. (Gulf Air)	GF
Hapag-Lloyd Fluggesellschaft MBH	HF
Horizon Midlands, Ltd.	HM
Indian Airlines Corp.	IC
Indonesia, Government of	QI
Intasun, LTD	WU
International Lease Finance Corp.	IE
Iran National Airlines Corp.	IR
Iraqi Airways	IA
ITEL Air, Ltd.	IU
Kuwait Airways Corporation	KU
LAN Chile Airlines (Lineas Aereas Nacional-Chile)	LA
Linea Aeran Del Cobra (LADECO)	UC
Linjeflyg A. B.	LF
Ling-Tempco-Vought, Inc.	Deleted
Luxair (Societe Anonyme Luxembourgeoise de Navigation Aeriennne)	LG
Maersk Air	DM
Malaysian Airline System Berhad	MH
Markair	ALK
Maritime Investment & Shipping Co., Ltd.	MI
Mey-Air A/S	Deleted
Monarch Airlines, Ltd.	MON
National Aeronautic & Space Administration (NASA)	NV
Nigeria Airways Ltd.	WT
Noga Export Import	NE
Nordair Ltd.	ND
Olympic Airways S.A.	OA
Orion Airways	OI
Owners Services Inc. (Midland Montagu Leasing Ltd.)	OJ

TABLE A

Airline Designator Codes
(Sheet 2)

AIRLINE	CODE
Pacific Southwest Airlines, Inc.	Deleted
Pacific Western Airlines, Ltd.	PW
Petrolair Systems Athens	PT
Piedmont Aviation, Inc.	PI
PLUNA (Primeras Lineas Uruguayas de Navegacion Aerea)	PLU
Polynesian Airways Ltd.	PH
Presidential	PRS
Quebcair	QB
Republic of Liberia	LU
Republique of Niger	AW
Royal Air Maroc (Compagnie Nationale de Transports Aeriens)	AT
Royal Brunei Airlines	BI
Royal Thai Air Force	TAF
Sabena Belgian World Airlines	SN
Saudi Arabian Airlines Corporation (Saudia)	SV
Servicio Aereo de Honduras S.A. (SAHSA)	SH
Servicios Aereos Cruzeiro do Sul, S.A. (Cruzeiro)	SC
Singapore Airlines Ltd. - SIA	SQ
Sobelair (Societe Belge de Transports par Air Societe Anonyme NV)	SE
South African Airways (SAA)	SA
Southwest Airlines Co., Ltd (Okinawa)	NU
Southwest Airlines, Inc.	AQ
Sudan Airways	SD
TAAG - (Transportes Aereos de Angola, S.A.R.L.)	DT
TACA International Airlines, S.A.	TA
Thai Airways, Ltd.	TH
Trans European Airways	TQ
Transair Ltd.	TZ
Transavia Holland B.V.	HV
Transportes Aereos Nacionales S.A. (TAN Airlines)	TX
Transportes Aereos Portugueses (TAP)	TP
Tunis Air (Societe Tunisienne de L'air)	TU
United Airlines, Inc.	UA
United States Air Force	US
US Air, Inc.	AL
Varig Airlines (Empresa de Viacao Aerea Rio Grandense)	RG
VASP (Viacao Aerea Sao Paulo, S.A.)	VP
Venezuela, Government of	VZ

TABLE A

Airline Designator Codes
(Sheet 3)

737
POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE	CODE
Western Air lines, Inc. (WAL)	WA
Wien Air Alaska	WE
Yemen Airways Corporation (North Yemen)	IY
Zambia Airways Corporation	QZ

TABLE A

Airline Designator Codes
(Sheet 4)

COMMON INSTALLATIONS	
USED ON ALL POWER PACK ASSEMBLIES - ALL POSITIONS	
INSTL PART NO.	IPL FIG. NO
65-19592-3 *[1]	1101
65-20996-1 *[1]	1103
65-25736-2	1107
65-45798-3 *[1]	1109
65-46851-1 *[1]	1111
65-46852-1	1112
65-46853-2	1113
65-50571-1 *[1]	1114
65-50882-1 *[1]	1115
65-52261-2 *[1]	1116
65-54476-1 *[1]	1118
65-55475-1 *[1]	1119
65-55749-1 *[1]	1120
65-56646-1 *[1]	1122
65-58122-1 *[1]	1123
65-58126-1	1125
65-58126-2	1125
65-58135-1	1126
65-58135-2 *[1]	1126
65-58135-3	1126
65-58135-4 *[1]	1126
65-58145-10 *[1]	1127
69-15935-1	1131
69-36670-1 *[1]	1134
69-37478-1	1135
69-45172-1	1136
69-50911-1 *[1]	1141

TABLE C - Common Installations

*[1] REFER TO APPLICABLE AIRLINE POWER PACK INSTALLATION MATRIX FOR SUPERSEDING INSTALLATION.

AIRLINE (AF)

MODEL 228

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-85379-3	1143	X	
65-19592-24 *[1]	1101		X	65-85379-4	1143		X
65-20996-19 *[1]	1103	X	X	69-15863-1	1130	X	X
65-21931-18	1114A	P	P	69-48936-1	1137	X	X
65-21931-22	1114A	P	P	69-50911-2 *[1]	1141	X	X
65-21950-5	1104	X		JT8D15ABG300 *[2]		P	P
65-21950-6	1104		X	JT8D15ABG314		P	P
65-22249-3	1105	X	X				
65-22454-8	1106	P	P				
65-22454-9	1106	P	P				
65-45798-4	1109	X	X				
65-45798-6	1109	X	X				
65-45798-7	1109	X	X				
65-45798-10 *[1]	1109	X	X				
65-45799-165	1109	X	X				
65-45799-170	1109	X	X				
65-46851-4 *[1]	1111	X	X				
65-50571-5 *[1]	1114	X	X				
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X				
65-55475-33 *[1]	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-2 *[1]	1122	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58135-47 *[1]	1126	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58234-15	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-70950-7	1138	X					
65-70950-8	1138		X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D-15 PER CR 02-005 AND 65C26425-1

AIRLINE (AH)				AIRLINE (AH)			
MODEL 2D6C				MODEL 2D6C			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-19684-1	1102	X	X	65-19684-1	1102	P	P
65-21950-3	1104	X	X	65-21950-2	1104	P	P
65-22249-3	1105	X	X	65-21950-3	1104	P	P
65-22454-8	1106	X	X	65-21950-5	1104	P	P
65-45798-1	1109	X	X	65-21950-6	1104		P
65-46851-4	1111	X	X	65-22249-1	1105	P	P
65-46853-1	1113	X	X	65-22249-3	1105	P	P
65-51534-1	1110	X	X	65-22454-8	1106	X	X
65-52261-1	1116	X	X	65-45798-1	1109	P	P
65-52971-1	1117	X	X	65-46851-4	1111	P	P
65-55750-1	1120	X		65-46853-1	1113	P	P
65-55750-2	1120		X	65-51534-1	1110	X	X
65-58125-3	1124	X	X	65-52261-1	1116	P	P
65-58135-18	1126	X		65-52971-1	1117	X	X
65-58135-20	1126		X	65-55750-1	1120	P	
65-58145-4	1127	X	X	65-55750-2	1120		P
65-58234-10	1128	X	X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-1	1142A	P	P	65-58145-4	1127	P	P
65-69659-10	1142A	P	P	65-58234-10	1128	P	P
65-70950-1	1138	P		65-58234-3	1128	X	X
65-70950-2	1138		P	65-65919-4	1108	X	X
65-70950-7	1138	P		65-69659-1	1142A	P	P
65-70950-8	1138		P	65-69659-10	1142A	P	P
69-15863-1	1130	X	X	65-70950-1	1138	P	
69-48936-1	1137	X	X	65-70950-2	1138		P
JT8D9BG35		P	P	65-70950-5	1138	X	
JT8D15BG42		P	P	65-70950-6	1138		X
				65-70950-7	1138	P	
				65-70950-8	1138		P
				65-85379-3	1143	P	
				65-85379-4	1143		P
				65-15863-1	1130	X	X
				65-48936-1	1137	X	X
				JT8D15BG35		P	P
				JT8D15BG42		P	P
				JT8D9BG21		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (AH)				AIRLINE (AH)			
MODEL 2D6				MODEL 2T4			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-22249-3	1105	X	X	65-19592-23*[1]	1101	X	
65-22454-8	1106	X	X	65-19592-24*[1]	1101		X
65-46851-4	1111	X	X	65-20996-1	1103	P	P
65-52971-1	1117	X	X	65-20996-21*[2]	1103	P	P
65-70950-7	1138	X		65-21950-5	1104	X	
65-70950-8	1138		X	65-21950-6	1104		X
69-15863-1	1130	X	X	65-22249-2	1105	X	X
JT8D15BG42		P	P	65-45798-10*[1]	1109	X	X
JT8D-17A		P	P	65-45799-148	1109	X	X
				65-45799-170	1109	P	P
				65-46851-4*[1]	1111	X	X
				65-51534-5	1110	X	X
				65-52971-1	1117	X	X
				65-54476-4	1118	X	X
				65-55749-9	1120	X	X
				65-58122-2	1123	X	X
				65-58125-3	1124	X	X
				65-58135-18	1126	X	
				65-58135-20	1126		X
				65-58135-30	1126	X	X
				65-58135-47	1126	X	X
				65-58234-3	1128	X	X
				65-65919-4	1108	X	X
				65-69659-13 *[3]	1142A	X	X
				65-70950 *[4]	1138	X	
				65-70950 *[4]	1138		X
				65-70950-11	1138	X	
				65-70950-12	1138		X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D17ABG301		P	P
				JT8D17RBG301BFE		P	P
				JT8D15 *[4]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPERATING INDICATION LIGHT PRESSURE SWITCH
PER RR 97031-55

*[3] ADDED PER RR 97031-25

*[4] ADDED PER RR 97031-24

*[5] ENGINE INTERMIX PER MC 3510-217K AND SB 71-1150

POWER PACK INSTALLATIONS
(AH)

AIRLINE (AL)

MODEL 2B7
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23*[1]	1101	X		65-70950-8	1138		X
65-19592-24*[1]	1101		X	65-85379-3	1143	X	
65-20996-19*[1]	1103	X	X	65-85379-4	1143		X
65-21931-18	1114A	X	X	69-28727-8	1133	X	X
65-21950-5	1104	X		69-48936-1	1137	X	X
65-21950-6	1104		X	69-50911-2*[1]	1141	X	X
65-22249-2	1105	X	X	JT8D15ABG308		P	P
65-45798-4	1109	X	X	JT8D15ABG311		P	P
65-45798-6	1109	X	X	JT8D-17*[2]		P	P
65-45798-7	1109	X	X	JT8D-17A*[2]		P	P
65-45798-10	1109	X	X	JT8D-17R*[2]		P	P
65-45799-1	1109	X	X	JT8D-17AR*[2]		P	P
65-46851-4*[1]	1111	X	X	JT8D-9A		P	P
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4*[1]	1118	X	X				
65-55475-33*[1]	1119	X	X				
65-55749-8*[1]	1120	X	X				
65-56646-2*[1]	1122	X	X				
65-58122-2*[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-47*[1]	1126	X	X				
65-58145-12*[1]	1127	X	X				
65-58234-5	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	P	P				
65-69659-2	1142A	P	P				
65-70950-7	1138	X					

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER MC 3510-176K AND SB 71-1108

*[3] ENGINE INTERMIX PER SB 737-71-1369

POWER PACK INSTALLATIONS
(AL)

AIRLINE (ALK)

 MODEL 2X6C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	ENGINE FIGURE	1	2
65-19592-23	*[1] 1101	X	
65-19592-24	*[1] 1101		X
65-20996-21	*[1] 1103	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-45798-4	1109	X	X
65-45798-5	1109	X	X
65-45798-10	1109	X	X
65-45799-170	1109	X	X
65-46851-5	*[1] 1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-54476-4	*[1] 1118	X	X
65-55749-9	*[1] 1120	X	X
65-58122-2	*[1] 1123	X	X
65-58125-3	1124	X	X
65-58126-20	1125	X	X
65-58135-21	1126	X	
65-58135-30	*[1] 1126	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-77451-4	1125	X	X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17ABG300		P	P
JT8D17ABG314		P	P
JT8D9	*[2]	P	P
JT8D9A	*[2]	P	P
JT8D15	*[2]	P	P
JT8D15A	*[2]	P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D17A PER SB 71-1139 AND MC 3510-211K

POWER PACK INSTALLATIONS

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (AMW)
 MODEL 277
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (AMW)
 MODEL 2U9
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-11	1103	X	X	65-21950-5	1104	X	
65-21931-18	1114A	X	X	65-21950-6	1104		X
65-21950-5	1104	X		65-27749-1	1105	X	X
65-21950-6	1104		X	65-45799-148	1109	X	X
65-22249-3	1105	X	X	65-46851-4	1111	X	X
65-22454-8	1106	X	X	65-51534-5	1110	X	X
65-45798-6	1109	X	X	65-58125-3	1124	X	X
65-45798-7	1109	X	X	65-58135-18	1126	X	
65-46851-4	1111	X	X	65-58135-20	1126		X
65-51534-8	1110	X	X	65-65919-4	1108	X	X
65-52971-1	1117	X	X	65-69659-1	1142A	X	X
65-55475-33	1119	X	X	65-70950-7	1136	X	
65-55749-8	1120	X	X	65-70950-8	1136		X
65-56646-2	1122	X	X	65-85379-3	1143	X	
65-58125-3	1124	X	X	65-85379-4	1143		X
65-58135-18	1126	X		69-15863-1	1130	X	X
65-58135-20	1126		X	69-48930-1	1137	X	X
65-58234-15	1128	X	X	69-56334-1	1139	X	X
65-65919-4	1108	X	X				
65-70950-7	1138	X		JT8D15BG300		P	P
65-70950-8	1138		X	JT8D9 *[1]		P	P
65-85379-3	1143	X		JT8D9A *[1]		P	P
65-85379-4	1143		X	JT8D15 *[1]		P	P
69-15863-1	1130	X	X	JT8D15A *[1]		P	P
69-21612-1	1132	X	X	JT8D17 *[1]		P	P
69-45172-3	1136	X	X				
69-48936-1	1137	X	X				
69-50911-2	1141	X	X				
JT8D15BG304		P	P				
JT8D9 *[1]		P	P				
JT8D9A *[1]		P	P				
JT8D15 *[1]		P	P				
JT8D15A *[1]		P	P				
JT8D17 *[1]		P	P				

*[1] ENGINE INTERMIX PER MC 7200MK3005

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (AMW)

AIRLINE (AN)
 MODEL 277
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-11	1103	X	X
65-21931-18	1114A	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-22454-8	1106	X	X
65-45798-6	1109	X	X
65-45798-7	1109	X	X
65-46851-4	1111	X	X
65-51534-8	1110	X	X
65-52971-1	1117	X	X
65-55475-33	1119	X	X
65-55749-8	1120	X	X
65-56646-2	1122	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-15	1128	X	X
65-65919-4	1108	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-21612-1	1132	X	X
69-45172-3	1136	X	X
69-48936-1	1137	X	X
69-50911-2	1141	X	X
JT8D15BG304		X	X

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (AP)
 MODEL 159
 ENGINE 1 65-73769-29
 ENGINE 2 65-73769-30

AIRLINE (AP)
 MODEL 293
 ENGINE 1 65-73769-9
 ENGINE 2 65-73769-10

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-3	1105	X	X	65-22249-2	1105	X	X
65-22454-15	1106	X	X	65-22454-8	1106	P	P
65-45798-1	1109	X	X	65-22454-9	1106	P	P
65-46853-1	1113	X	X	65-27808-1	1108	P	P
65-51534-1	1110	X	X	65-45798-1	1109	X	X
65-52261-1	1116	X	X	65-46853-1	1113	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55750-1	1120	X		65-52261-1	1116	X	X
65-55750-2	1120		X	65-52971-1	1117	X	X
65-57815-1	1140	X	X	65-55750-1	1120	X	
65-57815-2	1140	X	X	65-55750-2	1120		X
65-58125-1	1124	X	X	65-57815-1	1140	P	P
65-58135-12	1126	X	X	65-57815-2	1140	P	P
65-58145-4	1127	X	X	65-58125-1	1124	X	X
65-58234-3	1128	X	X	65-58135-12	1126	P	P
65-69659-1	1142A	X	X	65-58135-18	1126	P	
65-65919-4	1108	X	X	65-58135-20	1126		P
65-73778-1	(SB 78-1005)	X	X	65-58145-4	1127	X	X
69-15863-1	1130	X	X	65-58234-10	1128	P	P
69-48936-1	1137	X	X	65-58234-3	1128	X	X
JT8D7BG3 *[1]		X	X	65-65919-4	1108	P	P
JT8D9 *[1]				65-69659-1	1142A	X	X
JT8D9A *[1]				65-70950-1	1138	P	
				65-70950-2	1138		P
				65-70950-5	1138	P	
				65-70950-6	1138		P
				65-73778-1	(SB 78-1005)		
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D7BG3		X	X

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC 3510-199K AND SB 71-1123

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (AQ)
 MODEL 2H4
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (AQ)
 MODEL 2H4
 ENGINE 1 65-73769-61
 ENGINE 2 65-73769-62

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-3	1104	P	P	65-19684-1	1102	X	X
65-21950-5	1104	P		65-21950-3	1104	X	X
65-21950-6	1104		P	65-22249-2	1105	X	X
65-22249-2	1105	X	X	65-22454-8	1106	X	X
65-51534-1	1110	X	X	65-45798-1	1109	X	X
65-52971-1	1117	X	X	65-46853-1	1113	X	X
65-55475-28	1119	P	P	65-52261-1	1116	X	X
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58135-18	1126	X		65-55750-1	1121	X	
65-58135-20	1126		X	65-55750-2	1121		X
65-58145-12	1127	X	X	65-55750-2	1121		X
65-58234-10	1128	P	P	65-58125-1	1124	X	X
65-58234-3	1128	P	P	65-58135-18	1126	X	
65-58652-2	1112	P	P	65-58135-20	1126		X
65-65919-4	1108	X	X	65-58145-4	1127	X	X
65-69659-1	1142A	X	X	65-58234-10	1128	X	X
65-70950-5	1138	X		65-58234-3	1128	X	X
65-70950-6	1138		X	65-65919-4	1108	X	X
65-85379-3	1143	X		65-70950-1	1138	X	
65-85379-4	1143		X	65-70950-2	1138		X
69-15863-1	1130	P	P	65-70950-5	1138	X	
69-28727-8	1133	P	P	65-70950-6	1138		X
69-48936-1		X	X	69-15863-1	1130	X	X
JT8D9ABG42		P	P	69-48936-1	1137	X	X
JT8D9ABG301		P	P	JT8D9BG9		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

BOEING
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (AQ)
 MODEL 2H4
 ENGINE 1 65-73769-69
 ENGINE 2 65-73769-70

AIRLINE (AQ)
 MODEL 2H4C
 ENGINE 1 65-73769-71
 ENGINE 2 65-73769-72

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-2	1105	X	X
65-22454-8	1106	X	X	65-22454-8	1106	X	X
65-45798-1	1109	X	X	65-45798-1	1109	X	X
65-46853-1	1113	X	X	65-46853-1	1113	X	X
65-52261-1	1116	X	X	65-52261-1	1116	X	X
65-52971-1	1117	X	X	65-52971-1	1117	X	X
65-55750-1	1121	X		65-55750-1	1121	X	
65-55750-2	1121		X	65-55750-2	1121		X
65-58125-1	1124	X	X	65-58125-1	1124	X	X
65-58135-18	1126	X		65-58135-18	1126	X	
65-58135-20	1126		X	65-58135-20	1126		X
65-58145-4	1127	X	X	65-53145-4	1127	X	X
65-58234-10	1128	X	X	65-58234-10	1128	X	X
65-58234-3	1128	X	X	65-58234-3	1128	X	X
65-65919-4	1108	X	X	65-65919-4	1108	X	X
65-70950-1	1138	X		65-70950-1	1138	X	
65-70950-2	1138		X	65-70950-2	1138		X
65-70950-5	1138	X		65-70950-5	1138	X	
65-70950-6	1138		X	65-70950-6	1138		X
69-15863-1	1130	X	X	69-15863-1	1130	X	X
69-48936-1	1137	X	X	69-48936-1	1137	X	X
JT8D9BG3		X	X	JT8D9BG9		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN.
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (AQ)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (AQ)

MODEL 2H4

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19592-23	*[1] 1101	X	
65-19592-24	*[1] 1101		X
65-20996-1	*[2] 1103	P	P
65-20996-21	1103	P	P
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-45798-10	*[1] 1109	P	P
65-45798-4	*[1] 1109	P	P
65-45799-1	1109	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-54476-4	*[1] 1118	P	P
65-55475-28	*[1] 1119	X	X
65-55749-9	*[1] 1120	P	P
65-58122-2	*[1] 1123	P	P
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-47	*[1] 1126	P	P
65-58145-12	*[1] 1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D9ABG300		P	P
JT8D9ABG301		P	P
JT8D9ABG314		P	P

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC3540-11K AND SB 80-1018

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP SYMBOLS

SYMBOLS X= USED ON ALL ENGINES FOR POSITION SHOWN
P= PARTIAL USAGE

POWER PACK INSTALLATIONS

Dec 5/84

(AQ)

71-03-01
Page 1210A

737

POWERPLANT BUILDUP

BOEING

COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (AR)				AIRLINE (AR)			
MODEL 287				MODEL 287C			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-19684-1	1102	X	X
65-21950-2	1104	P	P	65-21950-2	1104	X	X
65-21950-3	1104	X	X	65-22249-2	1105	X	X
65-22249-1	1105	P	P	65-22454-8	1106	X	X
65-22249-2	1105	P	P	65-46853-1	1113	X	X
65-22454-8	1106	P	P	65-51534-1	1110	X	X
65-22454-9	1106	P	P	65-52261-1	1116	X	X
65-45798-1	1109	P	P	65-55750-1	1120	X	
65-46853-1	1113	P	P	65-55750-2	1120		X
65-50571-4	1114	P	P	65-58125-1	1124	X	X
65-51534-1	1110	X	X	65-58135-18	1126	X	
65-52261-1	1116	P	P	65-58135-20	1126		X
65-52971-1	1117	X	X	65-58145-4	1127	X	X
65-55450-1	1120	P		65-58234-10	1128	X	X
65-55750-2	1120		P	65-58234-3	1128	X	X
65-58125-1	1124	P	P	65-65919-4	1108	X	X
65-58125-3	1124	P	P	65-69659-10	1142A	X	X
65-58135-18	1126	X		65-70950-1	1138	X	
65-58135-20	1126		X	65-70950-2	1138		X
65-58145-4	1127	P	P	65-15863-1	1130	X	X
65-58234-10	1128	X	X	69-48936-1	1137	X	X
65-58234-3	1128	X	X	JT8D9BG9		X	X
65-65919-4	1108	X	X				
65-69659-1	1142A	P	P				
65-69659-10	1142A	P	P				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-85379-1	1140	P					
65-85379-2	1140		P				
65-85379-3	1140	P					
65-85379-4	1140		P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D9ABG21		P	P				
JT8D9BG21		P	P				
JT8D9BG3		P	P				
JT8D9BG35		P	P				
JT8D9BG9		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (AR)

Jan 5/81

71-03-01
 Page 1211

AIRLINE (ARK)

AIRLINE (AS)

MODEL 2E7
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

MODEL 290C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

		ENGINE				ENGINE	
INSTALLATION	FIGURE	1	2	INSTALLATION	FIGURE	1	2
65-19592-23*[1]	1101	X		65-21931-18	1114A	X	X
65-19592-24*[1]	1101		X	65-21950-5	1104	X	
65-20996-19*[1]	1103	X	X	65-21950-6	1104		X
	*[2]			65-22249-2	1105	X	X
65-21950-5	1104	X		65-45798-5	1109	X	X
65-21950-6	1104		X	65-45798-6	1109	X	X
65-22249-1	1105	X	X	65-45798-7	1109	X	X
65-45798-10*[1]	1109	X	X	65-51534-8	1110	X	X
65-45798-4*[1]	1109	X	X	65-46851-5	1111	X	X
65-45799-148	1109	X	X	65-52971-1	1117	X	X
65-45799-170	1109	X	X	65-55475-33	1119	X	X
65-46851-4*[1]	1111	X	X	65-55475-34	1119	X	X
65-51534-5	1110	X	X	65-55749-8	1120	X	X
65-52971-1	1117	X	X	65-56646-2	1122	X	X
65-54476-4*[1]	1118	X	X	65-58125-3	1124	X	X
65-55475-28*[1]	1119	X	X	65-58126-20	1125	X	X
65-55749-9*[1]	1120	X	X	65-58135-18	1126	X	X
65-58122-2*[1]	1123	X	X	65-58135-20	1126	X	X
65-58125-3	1124	X	X	65-58135-21	1126	X	
65-58135-18	1126	X	X	65-58135-30	1126	X	X
65-58135-20	1126		X	65-58145-12	1127	X	X
65-58135-30*[1]	1126	X	X	65-58234-17	1128	X	X
65-58135-47*[1]	1126	X	X	65-65919-4	1108	X	X
65-58145-12*[1]	1127	X	X	65-70950-11	1138	X	
65-58234-3	1128	X	X	65-70950-12	1138		X
65-65919-4	1108	X	X	65-77451-4	1125	X	X
65-69659-1	1142A	X	X	65-85379-3	1143	X	
65-70950-11	1138	X		65-85379-4	1143		X
65-70950-12	1138		X	69-28727-8	1133	X	X
65-85379-3	1143	X		69-36670-2	1134	X	X
65-85379-4	1143		X	69-48936-1	1137	X	X
69-28727-8	1133	X	X	69-50911-2	1141	X	X
69-48936-1	1137	X	X	JT8D17BG301		X	X
69-56334-1	1139	X	X				
JT8D17ABG300		X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH ADDED PER MC3540-11K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(ARK) (AS)

AIRLINE (AT)				AIRLINE (AW)			
MODEL 2B6				MODEL 2N9C			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		61-30196-001	1117	X	X
65-21950-6	1104		X	61-30198-193	1117	X	X
65-22249-2	1105	X	X	61-30200-001	1117	X	X
65-22249-3	1105	X	X	61-30404-3	1117	X	X
65-51534-1	1110	X	X	65-21950-5	1104	X	X
65-52971-1	1117	X	X	65-22249-1	1105	X	X
65-58125-3	1124	X	X	65-45799-147	1109	X	X
65-58135-18	1126	X		65-45799-148	1109	X	X
65-58135-20	1126		X	65-46851-5	1111	X	X
65-58234-3	1128	X	X	65-51534-1	1110	X	X
65-65919-4	1108	X	X	65-58125-3	1124	X	X
65-69659-1	1142A	X	X	65-58135-21	1126	X	
65-70950-7	1138	X		65-58135-26	1126	X	X
65-70950-8	1138		X	65-58234-3	1128	X	X
65-85379-3	1143	X		65-65919-4	1108	X	X
65-85379-4	1143		X	65-69659	1142A	X	X
69-15863-1	1130	X	X	65-70950-11	1138	X	
69-48936-1	1137	X	X	65-70950-12	1138		X
JT8D15BG42 *[1]		X	X	65-85379-3	1143	X	
JT8D-15A *[1]				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D17BG301		X	X

*[1] ENGINE INTERMIX PER MC 7200MK3031 AND SERVICE BULLETIN 737-71-1391.

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(AT) (AW)

Mar 1/01

71-03-01
Page 1212A

AIRLINE (BA)

MODEL 236

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		69-28727-8	1133	X	X
65-19592-24 *[1]	1101		X	69-48936-1	1137	X	X
65-21950-5	1104	X		69-45172-3	1136	X	X
65-20996-21 *[1]	1103	X	X	JT8D15BG300 *[2]		P	P
65-21950-6	1104		X	JT8D15BG301 *[2]		P	P
65-22249-3	1105	X	X	JT8D15ABG300		P	
65-25736-3	1107	X	X	JT8D15ABG314		P	P
65-45798-4	1109	X	X	JT8D15A *[2]*[3]		P	P
65-45798-6	1109	X	X	JT8D15 *[3]		P	P
65-45798-8	1109	X	X	JT8D17A *[3]		P	P
65-45799-165	1109	X	X				
65-45799-170	1109	X	X				
65-46851-4 *[1]	1111	X	X				
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-55475-28	1119	X	X				
65-58125-3	1124	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58145-12	1127	X	X				
65-58234-16	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-69659-2	1142A	X	X				
65-70950-7	1138	X					
65-70950-8	1138		X				
65-85379-3	1143	X					
65-85379-4	1143		X				
69-21612-1	1132	X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER MC 3510-207K AND SB 71-1127

*[3] ENGINE INTERMIX PER MC 3510-231K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (BAT)

MODEL 236
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19592-3	1101	P	P
65-19592-23 *[1]	1101	P	
65-19592-24 *[1]	1101		P
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-25736-3 *[1]	1107	X	X
65-45798-6	1109	X	X
65-45798-8	1109	X	X
65-45799-165	1109	X	X
65-46851-4 *[1]	1111	X	X
65-51534-8	1110	X	X
65-52971-1	1117	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12 *[1]	1127	X	X
65-58234-16	1128	X	X
65-65919-4	1108	X	X
65-69659-2	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-21612-1	1132	X	X
69-28727-8	1133	X	X
69-45172-3 *[1]	1136	X	X
69-48936-1	1137	X	X
JT8D15BG300 *[2]		P	P
JT8D15BG301 *[2]		P	P
JT8D15A *[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER MC 3510-207K AND SB 71-1127

737 **BOEING**
POWERPLANT BUILDUP **COMMERCIAL JET**
OVERHAUL MANUAL

AIRLINE (BB)
 MODEL 2R4C
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-1 *[2]	1103	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-45798-5	1109	X	X
65-45799-147	1109	X	X
65-45799-148	1109	X	X
65-46851-5 *[1]	1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58126-20	1125	X	X
65-58135-20	1126		X
65-58135-21	1126	X	
65-58135-30 *[1]	1126	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-77451-4	1125	X	X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D15BG306		X	X

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER 3540-11K AND SB 80-1022

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (BI)			AIRLINE (BI)				
MODEL 2M6C ENGINE 1 65-73769-23 ENGINE 2 65-73769-24			MODEL 2M6 ENGINE 1 65-73769-51 ENGINE 2 65-73769-52				
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-1	1105	X	X
65-22249-1	1105	X	X	65-22454-8	1106	X	X
65-46851-4	1111	X	X	65-46851-4	1111	X	X
65-51534-5	1110	X	X	65-51534-5	1110	X	X
65-52971-1	1117	X	X	65-52971-1	1117	X	X
65-58125-3	1124	X	X	65-58125-3	1124	X	X
65-58135-18	1126	X		65-58135-18	1126	X	
65-58135-20	1126		X	65-58135-20	1126		X
65-58234-3	1128	X	X	65-58234-9	1128	X	X
65-65919-4	1108	X	X	65-58234-12	1128	X	X
65-69659-1	1142A	X	X	65-65919-4	1108	X	X
65-70950-7	1138	X		65-69659-1	1142A	X	X
65-70950-8	1138		X	65-70950-7	1138	X	
65-85379-3	1143	X		65-70950-8	1138		X
65-85379-4	1143		X	65-85379-3	1143	X	
69-15863-1	1130	X	X	65-85379-4	1143		X
69-48936-1	1137	X	X	69-15863-1	1130	X	X
JT8D15BG301*[1]		P	P	69-48936-1	1137	X	X
JT8D-15A*[1]		P	P	JT8D15BG42*[1]		P	P
				JT8D-15A*[1]		P	P

*[1] ENGINE INTERMIX PER MC 3510-212K AND SB 71-1141

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (BL)
 MODEL 2K9
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-22249-2	1105	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-57815	1140	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17BG301 *[1]*[2]		P	P
JT8D9 *[1]*[2]		P	P
JT8D9A *[1]*[2]		P	P
JT8D15 *[2]		P	P

*[1] ENGINE INTERMIX PER MC 3510-145K AND SB 71-1089

*[2] ENGINE INTERMIX PER MC 3510-20K AND SB 71-1119

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (BU)
MODEL 205, 205C
ENGINE 1 65-73769-15
ENGINE 2 65-73769-16

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65C13071-1	1130	P	P
65-20996-1 *[3]	1103	P	P	69-15863-1	1130	X	X
65-20996-11 *[3]	1103	P	P	P69-21612-1	1132	P	P
65-21950-3	1104	P	P	P69-48936-1	1137	X	X
65-21950-5	1104	P		P69-60143-1	1142	P	P
65-21950-6	1104		P	PJT8D17BG301		P	P
65-22249-2	1105	X	X	JT8D17BG42		P	P
65-22454-8	1106	P	P	JT8D9ABG42 *[1]		P	P
65-22454-9	1106	P	P	JT8D9BG21 *[1]		P	P
65-45798-1	1109	P	P	JT8D9BG3 *[1]		P	P
65-46853-1	1113	P	P	JT8D9BG9		P	P
65-51534-1	1110	X	X	JT8D9ABG301 *[2]		P	P
65-52261-1	1116	P	P	JT8D9 *[4]		P	P
65-52971-1	1117	X	X	JT8D9A *[4]		P	P
65-55750-1	1121	P		JT8D15 *[4]		P	P
65-55750-2	1121		P	JT8D15A *[4]		P	P
65-58125-1	1124	P	P	JT8D17 *[4]		P	P
65-58125-3	1124	P	P	JT8D17A *[4]		P	P
65-58135-12	1126	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58145-14	1126	P	P	*[1] ENGINE INTERMIX WITH JT8D-17 PER MC3510-68, MC3510-70K			
65-58145-4	1127	P	P				
65-58234-10	1128	P	P				
65-58234-10	1128	X	X	*[2] ENGINE INTERMIX WITH JT8D- 17BG301 PER MC 3510-82			
65-65919-4	1108	X	X				
65-69659-1	1142A	P	P				
65-69659-2	1142A	P	P	*[3] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC3540-11K AND SB 80-1026			
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-70950-11	1138	P		*[4] ENGINE INTERMIX PER MC 3510-235K AND SB 71-1182			
65-70950-12	1138		P				
65-85379-3	1143	P					
65-85379-4	1143		P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (BU)

MODEL 205

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-70950-12	1138		X
65-19592-24 *[1]	1101		X	65-85379-3	1143	X	
65-20996-24	1103	X	X	65-85379-4	1143		X
65-21931-23	1114A	X	X	69-28727-8	1133	X	X
65-21950-5	1104	X		69-48936-1	1137	X	X
65-21950-6	1104		X	69-50911-2 *[1]	1141	X	X
65-22249-3	1105	X	X	JT8D17ABG314		X	X
65-45798-3	1109	X	X				
65-45798-4	1109	X	X				
65-45798-7	1109	X	X				
65-45798-10	1109	X	X				
65-45799-148	1109	X	X				
65-45799-166	1109	X	X				
65-46851-4 *[1]	1111	X	X				
65-51534-5	1110	X	X				
65-52971-1	1117	X	X				
65-55475-33 *[1]	1119	X	X				
65-54476-4 *[1]	1118	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-3 *[1]	1122	X	X				
65-58125-3	1124	X	X				
65-58126-24	1125	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-47 *[1]	1126	X	X				
65-58234-19	1128	X	X				
65-65919-4	1108	X	X				
65-69659-2	1142A	X	X				
65-70950-11	1138	X					

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (BU)				AIRLINE (CA)			
MODEL 2A6				MODEL 2T4			
ENGINE 1 65-73769-69				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-70				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19592-23 *[1]	1101	X	
65-20996-1 *[2]	1103	X	X	65-19592-24 *[1]	1101		X
65-21950-3	1104	X	X	65-21950-5	1104	X	
65-22249-2	1105	X	X	65-21950-6	1104		X
65-22454-8	1106	X	X	65-22249-2	1105	X	X
65-45798-1	1109	X	X	65-45798-10 *[1]	1109	X	X
65-46853-1	1113	X	X	65-45799-148	1109	X	X
65-51534	1110	X	X	65-45799-170	1109	X	X
65-52261-1	1116	X	X	65-46851-4	1111	X	X
65-52971-1	1117	X	X	65-51534-5	1110	X	X
65-55750-1	1120	X		65-52971-1	1117	X	X
65-55750-2	1120		X	65-54476-4	1118	X	X
65-58125-1	1124	X	X	65-55475-28 *[1]	1119	X	X
65-58135-18	1126	X		*[3]			
65-58135-20	1126		X	65-55749-9	1120	X	X
65-58145-4	1127	X	X	65-58122-2	1123	X	X
65-58234-10	1128	X	X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-2	1142A	X	X	65-58135-30	1126	X	X
65-70950-1	1138	X		65-58135-47	1126	X	X
65-70950-2	1138		X	65-58145-12 *[1]	1127	X	X
69-15863-1	1130	X	X	*[3]			
69-48936-1	1137	X	X	65-58234-3	1128	X	X
JT8D9BG9		P	P	65-65919-4	1108	X	X
JT8D9 *[4]		P	P	65-69659-1	1142A	X	X
JT8D9A *[4]		P	P	65-70950-11	1138	X	
JT8D15 *[4]		P	P	65-70950-12	1138		X
JT8D15A *[4]		P	P	65-85379-3	1143	X	
JT8D17 *[4]		P	P	65-85379-4	1143		X
JT8D17A *[4]		P	P	69-28727-8 *[3]	1133	X	X
				69-48936-1	1137	X	X
				JT8D17ABG301		X	X

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE STARTER VALVE OPERATING INDICATION LIGHT PRESSURE SWITCH ADDED PER MC3540-14K

*[3] ADDED PER RR 97096-2, & RR 97096-27

*[4] ENGINE INTERMIX PER MC 3510-235K AND SB 71-1182

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



AIRLINE (CA)
 MODEL 2T4C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-58126-20	1125	X	X
65-19592-24	*[1] 1101		X	65-58135-21	1126	X	
65-20996-19	*[1] 1103	X	X	65-58135-30	*[1] 1126	X	X
65-21950-5	1104	X		65-58135-47	*[1] 1126	X	X
65-21950-6	1104		X	65-58145-12	*[1] 1127	X	X
65-22249-2	1105	X	X	65-58234-3	1128	X	X
65-45798-4	1109	X	X	65-65919-4	1108	X	X
65-45798-5	1109	X	X	65-69659-1	1142A	X	X
65-45798-10	1109	X	X	65-70950-11	1138	X	
65-45799-150	1109	X	X	65-70950-12	1138		X
65-46851-5	*[1] 1111	X	X	65-77451-4	1125	X	X
65-51534-5	1110	X	X	65-85379-3	1143	X	
65-52971-1	1117	X	X	65-85379-4	1143		X
65-54476-4	*[1] 1118	X	X	69-28727-8	1133	X	X
65-55475-28	*[1] 1119	X	X	69-48936-1	1137	X	X
65-55749-9	*[1] 1120	X	X	69-56334-1	1139	X	X
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X	JT8D17ABG300		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (CI)

MODEL 209

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	P		65-58135-1	1126	X	X
65-19592-24 *[1]	1101		P	65-58135-2	1126	X	X
65-19592-29 *[1]	1101	P		65-58135-3	1126	X	X
65-19592-30 *[1]	1101		P	65-58135-47 *[1]	1126	X	X
65-20996-24 *[1]	1103	X	X	65-58145-10	1127	X	X
65-21931-23	1114A	X	X	65-58234-19	1128	X	X
65-21950-5	1104	X		65-65919-4	1108	X	X
65-21950-6	1104		X	65-69659-2 *[2]	1142A	X	X
65-22249-2	1105	X	X	65-70950-5	1138	X	
65-25736-2	1107	X	X	65-70950-6	1138		X
65-45798-3	1109	X	X	65-85379-3	1143	X	
65-45798-4	1109	X	X	65-85379-4	1143		X
65-45798-7	1109	X	X	69-15863-1	1130	X	X
65-45798-10	1109	X	X	69-15935-1	1131	X	X
65-45799-166	1109	X	X	69-36670-1	1134	X	X
65-46851-1	1111	X	X	69-37478-1	1135	X	X
65-46852-1	1112	X	X	69-45172-1	1136	X	X
65-46853-2	1113	X	X	69-48936-1	1137	X	X
65-46858-131	1138	X		69-50911-2 *[1]	1141	X	X
65-46858-132	1138		X				
65-50571-5 *[1]	1114	X	X	JT8D9ABG314		X	X
65-50882-1	1115	X	X				
65-51534-5	1110	X	X				
65-52261-2	1116	X	X				
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X				
65-55475-33 *[1]	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-3 *[1]	1122	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-24	1125	X	X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
*[2] METALCAL INSTALLATION PER RR97126-2

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (CI)				AIRLINE (CP)			
MODEL 281				MODEL 217			
ENGINE 1 65-73769-67				ENGINE 1 65-73769-41			
ENGINE 2 65-73769-68				ENGINE 2 65-73769-42			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-20996-1*[1]	1103	X	X	65-21950-3	1104	X	X
65-21950-3	1104	X	X	65-22249-3	1105	X	X
65-22249-2	1105	X	X	65-22454-9	1105	X	X
65-22454-16	1106	X	X	65-27808-1	1108	P	P
65-45798-1	1103	X	X	65-45798-1	1109	X	X
65-45799-1	1109	X	X	65-46853-1	1113	X	X
65-48653-1	1113	X	X	65-52261-1	1116	X	X
65-51534-1	1110	X	X	65-52971-1	1117	X	X
65-52261-1	1116	X	X	65-55750-1	1120	X	
65-52971-1	1117	X	X	65-55750-2	1120		X
65-55750-1	1121	X	X	65-58125-1	1124	X	X
65-58125-1	1124	X	X	65-58135-12	1126	P	P
65-58135-18	1126	X		65-58135-18	1126	P	
65-58135-20	1126		X	65-58135-20	1126		P
65-58145-4	1127	X	X	65-58145-1	1127	X	X
65-58234-10	1128	X	X	65-58145-4	1127	X	X
65-65919-4	1108	X	X	65-58234-10	1128	P	P
65-69659-2	1142A	X	X	65-58234-3	1128	X	X
65-70950-5	1138	X		65-65919-4	1108	P	P
65-70950-6	1138		X	65-70950-1	1138	P	
69-15863-1	1130	X	X	65-70950-2	1138		P
69-48936-1	1137	X	X	65-70950-5	1138	P	
JT8D78G3		P	P	65-70950-6	1138		P
JT8D7BG9		P	P	65-73778-1	(SB 78-1005)	P	P
				69-28727-1	1133	X	X
				69-48936-1	1137	X	X
				69-56334-1	1139	X	X
				JT8D9BG8		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC3540-11K AND SB 80-1012

OVERHAUL MANUAL

AIRLINE (CP)			AIRLINE (CP)				
MODEL 217			MODEL 217				
ENGINE 1 65-73769-51			ENGINE 1 65-73769-51				
ENGINE 2 65-73769-52			ENGINE 2 65-73769-52				
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-19592-23 *[1]	1101	X	
65-21950-6	1104		X	65-19592-24 *[1]	1101		X
62-22249-3	1105	X	X	65-21950-5	1104	X	
65-22454-9	1106	X	X	65-21950-6	1104		X
65-46851-4	1111	X	X	65-22249-3	1105	X	X
65-52971-1	1117	X	X	65-22454-9	1106	X	X
65-55475-28	1119	X	X	65-45798-10 *[1]	1109	X	X
65-58125-3	1124	X	X	65-45799-148	1109	X	X
65-58135-18	1126	X		65-46851-4 *[1]	1111	X	X
65-58135-20	1126		X	65-51534-5	1110	X	X
65-58135-30	1126	X	X	65-52971-1	1117	X	X
65-58145-15	1127	X	X	65-54476-4 *[1]	1118	X	X
65-58234-3	1128	X	X	65-55475-28 *[1]	1119	X	X
65-65919-4	1108	X	X	65-55749-9 *[1]	1120	X	X
65-70950-11	1138	X		65-58122-2 *[1]	1123	X	X
65-70950-12	1138		X	65-58125-3	1124	X	X
65-85379-3	1143	X		65-58135-18	1126	X	
65-85379-4	1143		X	65-58135-20	1126		X
65C13071-3	1130	X	X	65-58135-30	1126	X	X
69-28727-8	1133	X	X	65-58135-47 *[1]	1126	X	X
69-48936-1	1137	X	X	65-58145-15 *[1]	1127	X	X
69-56334-1	1139	X	X	65-58234-3	1128	X	X
JT8D17BG304 *[2]		P	P	65-65919-4	1108	X	X
JT8D17A *[2]		P	P	65-69659-1	1142A	X	X
JT8D15 *[2]		P	P	65-70950-11	1138	X	
JT8D9A *[2]		P	P	65-70950-12	1138		X
JT8D9 *[2]		P	P	65-85379-3	1143	X	X
				65C13071-3	1130	X	X
				69-28727-8	1133	X	X
				69-48936-1	1137	X	X
				69-56334-1	1139	X	X
				JT8D17ABG304 *[3]		P	P
				JT8D17 *[3]		P	P
				JT8D15 *[3]		P	P
				JT8D9A *[3]		P	P
				JT8D9 *[3]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D-17A PER MC 3510-149K & SB 71-1082 AND INTERMIX WITH JT8D-9, -9A, -15 PER CR 5-225 & 65C15686-20

*[3] ENGINE INTERMIX WITH JT8D-17 PER MC 3510-119 & 65C25674-5, AND INTERMIX WITH JT8D-9, -9A, -15 PER CR 5-225 & 65C25674-5 AND, MC 3510-183K & SB 71-1117

POWER PACK INSTALLATIONS
(CP)

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (DF)				AIRLINE (DL)			
MODEL 230				MODEL 232			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-11	1103	X	X	65-19592-23 *[1]	1101	X	
65-21931-18	1114A	X	X	65-19592-24 *[1]	1101		X
65-21950-5	1104	X		65-20996-22 *[1]	1103	X	X
65-21950-6	1104		X	65-20996-19 *[1]	1103	X	X
65-22249-3	1105	X	X	65-21931-21	1114A	X	X
65-25736-3	1107	X	X	65-21950-5	1104	X	
65-45798-6	1109	X	X	65-21950-6	1104		X
65-45798-7	1109	X	X	65-22249-2	1105	X	X
65-46851-4	1111	X	X	65-45798-4	1109	X	X
65-51534-5	1110	X	X	65-45798-6	1109	X	X
65-52971-1	1117	X	X	65-45798-7	1109	X	X
65-55475-28	1119	X	X	65-45798-10	1109	X	X
65-55749-8	1120	X	X	65-45799-1	1109	X	X
65-56646-2	1114	X	X	65-46851-4 *[1]	1111	X	X
65-57815	1140	X	X	65-51534-8	1110	X	X
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58135-18	1126	X		65-54476-4 *[1]	1118	X	X
65-58135-20	1126		X	65-55475-33 *[1]	1119	X	X
65-58145-12	1127	X	X	65-55749-8 *[1]	1120	X	X
65-58234-3	1128	X	X	65-56646-3 *[1]	1122	X	X
65-58234-15	1128	X	X	65-58122-2 *[1]	1123	X	X
65-65919-4	1108	X	X	65-58125-3	1124	X	X
65-69659-1	1142A	X	X	65-58135-18	1126	X	
65-70950-23	1138	X		65-58135-20	1126		X
65-70950-24	1138		X	65-58135	1126	X	X
65-85379-3	1143	X		65-58145-12	1127	X	X
65-85379-4	1143		X	65-58234-20	1128	X	X
69-21612-1	1132	X	X	65-65919-4	1108	X	X
69-28727-8	1133	X	X	65-69659-1	1142A	X	X
69-48936-1	1137	X	X	65-69659-2	1142A	X	X
69-50911-2	1141	X	X	65-70950-7	1138	X	
JT8D15BG309		X	X	65-70950-8	1138		X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-28727-8	1133	X	X
				69-37478-2	1135	X	X
				69-48936-1	1137	X	X
				69-50911-2	1141	X	X
				JT8D15ABG315		X	X

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (DM)
 MODEL 2L9
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-1	1105	X	X
65-46851-4	1111	X	X
65-51534-1	1110	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	X
65-58135-20	1126	X	X
65-58135-26	1126	X	X
65-58145-12	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	P	
65-70950-12	1138		P
65-70950-5	1138	P	
65-70950-6	1138		P
65-70950-7	1138	P	
65-70950-8	1138		P
65-85379-3	1143	X	
65-85379-4	1143	X	
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D17BG301*[1]		P	P
JT8D17BG42*[1]		P	P

*[1] ENGINE INTERMIX WITH JT8D-9, -9A, OR -15 PER MC 3510-102K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (DT)
 MODEL 200C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

AIRLINE (DT)
 MODEL 2M2
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[2]	1103	X	X	65-19592-23 *[1]	1101	P	
65-21950-5	1104	X		65-19592-24 *[1]	1101		P
65-21950-6	1104		X	65-20996-1 *[2]	1103	P	P
65-22249-1	1105	X	X	65-20996-21 *[1]	1103	P	P
65-22454-8	1106	X	X	65-20996-24 *[1]	1103	P	P
65-45798-5	1109	X	X	65-21950-5	1104	X	
65-45799-147	1109	X	X	65-21950-6	1104		X
65-45799-148	1109	X	X	65-22249-1	1105	X	X
65-46851-5	1113	X	X	65-45798-4	1109	P	P
65-51534-1	1110	X	X	65-45798-5	1109	X	X
65-52971-1	1117	X	X	65-45798-10	1109	P	P
65-58125-3	1124	X	X	65-45799-170	1109	P	P
65-58126-20	1125	X	X	65-46851-5 *[1]	1111	X	X
65-58135-18	1126	X		65-51534-5	1110	X	X
65-58135-20	1126		X	65-52971-1	1117	X	X
65-58135-21	1126	X		65-54476-4 *[1]	1118	P	P
65-58135-26	1126	X	X	65-55749-9 *[1]	1120	P	P
65-58234-10	1128	X	X	65-58122-2 *[1]	1123	P	P
65-58234-3	1128	X	X	65-58125-3	1124	X	X
65-65919-4	1108	X	X	65-58126-20	1125	X	X
65-69659-1	1142A	X	X	65-58135-20	1126		X
65-70950-11	1138	X		65-58135-21	1126	X	
65-70950-12	1138		X	65-58135-30 *[1]	1126	X	X
65-77451-3	1125	X	X	65-58135-47 *[1]	1126	P	P
65-85379-3	1143	X		65-65919-4	1108	X	X
65-85379-4	1143		X	65-69659-1	1142A	X	X
69-15863-1	1130	X	X	65-70950-11	1138	X	
69-48936-1	1137	X	X	65-70950-12	1138		X
JT8D17BG42		X	X	65-77451-4	1125	X	X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D17BG301*[3]		P	P
				JT8D17ABG300*[3]		P	P

- *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
- *[2] INCLUDES ENGINE STARTER VALVE OPERATING INDICATION LIGHT PRESSURE SWITCH PER SB 80-1005
- *[3] ENGINE INTERMIX PER MC 3510-146K AND SB 71-1146

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (DY)				AIRLINE (EB)			
MODEL 2R4C				MODEL 2Q8			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-21950-5	1104	X	
65-21950-6	1104		X	65-21950-6	1104		X
65-22249-2	1105	X	X	65-22249-1	1105	X	X
65-45798-5	1109	X	X	65-46851-4	1111	X	X
65-46851-5	1111	X	X	65-51534-5	1110	X	X
65-51534-5	1110	X	X	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-58125-3	1124	X	X
65-58125-3	1124	X	X	65-58135-18	1126	X	
65-58126-20	1125	X	X	65-58135-20	1126		X
65-58135-20	1126		X	65-65919-4	1108	X	X
65-58135-21	1126	X		65-70950-7	1138	X	
65-58135-30	1126	X	X	65-70950-8	1138		X
65-65919-4	1108	X	X	65-85379-3	1143	X	
65-69659-1	1142A	X	X	65-85379-4	1143		X
65-70950-11	1138	X		69-15863-1	1130	X	X
65-70950-12	1138		X	69-48936-1	1137	X	X
65-70950-5	1138	X		JT8D15BG301		X	X
65-70950-6	1138		X				
65-77451-4	1125	X	X				
65-85379-3	1143	X					
65-85379-4	1143		X				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D17BG306		X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (EF)
 MODEL 247
 ENGINE 1 65-73769-7
 ENGINE 2 65-73759-8

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-3	1105	X	X
65-22454-7	1106	X	X
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-50882-2	1115	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55475-21	1119	X	X
65-55750-1	1120	X	
65-55750-2	1120		X
65-58125-1	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-4	1127	X	X
65-58234-10	1128	X	X
65-65919-4	1108	X	X
65-69659-7	1142A	X	X
65-70950-1	1138	X	
65-70950-2	1138		X
69-15863-1	1130	X	X
69-21612-1	1132	X	X
69-48936-1	1137	X	X
JT8D9BG4		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (EF)
MODEL 2Q8
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (EF)
MODEL 222
ENGINE 1 65-73769-5
ENGINE 2 65-73769-6

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1 *[1]	1103	X	X	65-19684-1	1102	X	X
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-1	1105	X	X
65-22249-2	1105	X	X	65-22454-8	1106	X	X
65-51534-1	1110	P	P	65-25736-3	1107	X	X
65-51534-5	1110	P	P	65-45798-1	1109	X	X
65-52971-1	1117	X	X	65-46853-1	1113	X	X
65-58125-3	1124	X	X	65-50571-2	1114	X	X
65-58135-18	1126	X		65-52261-1	1116	X	X
65-58135-20	1126		X	65-52971-1	1117	X	X
65-65919-4	1108	X	X	65-55750-1	1120	X	
65-69659-1	1142A	X	X	65-55750-2	1120		X
65-70950-5	1138	X		65-58125-1	1124	X	X
65-70950-6	1138		X	65-58135-18	1126	X	
65-85379-3	1143	X		65-58135-20	1126		X
65-85379-4	1143		X	65-58145-4	1127	X	X
69-15863-1	1130	X	X	65-58234-11	1128	X	X
69-48936-1	1137	X	X	65-58234-9	1128	X	X
JT8D9ABG301				65-65919-4	1108	X	X
				65-51534-1	1110	X	X
				65-70950-1	1138	X	
				65-70950-2	1138		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D7BG3		X	X

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC3540-11K AND SB 80-1034

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (EF)
MODEL 27A
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-58135-47	*[1] 1126	X	X
65-19592-24	1101		X	65-58145-10	1127	X	X
65-20996-24	*[1] 1103	X	X	65-58234-19	1128	X	X
65-21931-23	1114A	X	X	65-65919-4	1108	X	X
65-21950-5	1104	X		65-69659-1	1142A	X	X
65-21950-6	1104		X	65-70950-5	1138	X	
65-22249-2	1105	X	X	65-70950-6	1138		X
65-25736-2	1107	X	X	65-85379-3	1143	X	
65-45798-3	1109	X	X	65-85379-4	1143		X
65-45798-4	1109	X	X	69-15863-1	1130	X	X
65-45798-7	1109	X	X	69-15935-1	1131	X	X
65-46798-10	1109	X	X	69-36670-1	1134	X	X
65-45799-166	1109	X	X	69-37478-1	1135	X	X
65-46851-1	1111	X	X	69-45172-1	1136	X	X
65-46852-1	1112	X	X	69-48936-1	1137	X	X
65-46853-2	1113	X	X	69-50911-2	*[1] 1141	X	X
65-46858-131	1138	X		JT8D9 ABG 300		X	X
65-46858-132	1138		X				
65-50882-1	1115	X	X				
65-51534-5	1110	X	X				
65-52261-2	1116	X	X				
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	X	X				
65-55475-33	*[1] 1119	X	X				
65-55749-8	*[1] 1120	X	X				
65-56646-3	*[1] 1122	X	X				
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-24	1125	X	X				
65-58135-1	1126	X	X				
65-58135-2	1126	X	X				
65-58135-3	1126	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

POWER PACK INSTALLATIONS
(EF)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (EL)				AIRLINE (ET)			
MODEL 2S5C				MODEL 260			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19592-23 *[1]	1101	P	
65-21950-5	1104	X		65-19592-24 *[1]	1101		P
65-21950-6	1104		X	65-19592-29 *[1]	1101	P	
65-22249-3	1105	X	X	65-19592-30 *[1]	1101		P
65-45798-5	1109	X	X	65-20996-24 *[1]	1103	X	X
65-46851-5	1111	X	X	65-21931-21	1114A	X	X
65-51534-5	1110	X	X	65-22249-1	1105	X	X
65-52261-1	1116	X	X	65-25736-2	1107	X	X
65-52971-1	1117	X	X	65-45798-3	1109	X	X
65-55475-28	1119	X	X	65-45798-4	1109	X	X
65-58126-20	1125	X	X	65-45798-5	1109	X	X
65-58135-30	1126	X	X	65-45798-7	1109	X	X
65-58135-21	1126	X		65-46798-10	1109	X	X
65-58145-15	1127	X	X	65-46799-148	1109	X	X
65-58234-10	1128	X	X	65-46799-150	1109	X	X
65-58234-3	1128	X	X	65-46799-166	1109	X	X
65-65919-4	1108	X	X	65-46851-5 *[1]	1111	X	X
65-69659-1	1142A	X	X	65-46852-1	1112	X	X
65-70950-11	1138	X		65-46853-2	1113	X	X
65-70950-12	1138		X	65-46858-131	1138	X	
65-77451-4	1125	X	X	65-46858-132	1138		X
65C13071-3	1133	X	X	65-50882-1	1115	X	X
69-28727-8	1133	X	X	65-51534-5	1110	X	X
69-56334-1	1139	X	X	65-52261-2	1116	X	X
JT8D17BG300		P	P	65-52971-1	1117	X	X
JT8D15 *[1]		P	P	65-54476-4 *[1]	1118	X	X
JT8D9/9A *[1]		P	P	65-55475-28 *[1]	1119	X	X
				65-55475-33 *[1]	1119	X	X
				65-55749-8 *[1]	1120	X	X
				65-56646-3 *[1]	1122	X	X
				65-58122-2 *[1]	1123	X	X
				65-58125-3	1124	X	X
				65-58126-1	1125	X	X
				65-58126-2	1125	X	X

(Continued on next page)

*[1] ENGINE INTERMIX PER MC3510-88

INSTALLATIONS LISTED IN TABLE C MUST-ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (ET)(Continued)				AIRLINE (EZ)			
MODEL 260				MODEL 2A6			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-69			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-71			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-58126-20	1125	X	X	65-19684-1	1102	X	X
65-58126-24	1125	X	X	65-21950-3	1104	X	X
65-58135-1	1126	X	X	65-22249-2	1105	X	X
65-58135-3	1126	X	X	65-22454-8	1106	X	X
65-58135-20	1126		X	65-45798-1	1109	X	X
65-58135-21	1126	X		65-46853-1	1113	X	X
65-58135-30 *[1]	1126	X	X	65-51534-1	1110	X	X
65-58135-47 *[1]	1126	X	X	65-52261-1	1116	X	X
65-58145-12 *[1]	1127	X	X	65-52971-1	1117	X	X
65-58234-19	1128	X	X	65-55750-1	1120	X	
65-65919-4	1108	X	X	65-55750-2	1120		X
65-69659-1	1142A	X	X	65-58125-1	1124	X	X
65-70950-11	1138	X		65-58135-18	1126	X	
65-70950-12	1138		X	65-58135-20	1126		X
65-77451-4	1125	X	X	65-58145-4	1127	X	X
65-85379-3	1143	X		65-58234-10	1128	X	X
65-85379-4	1143		X	65-58234-3	1128	X	X
69-15935-1	1131	X	X	65-65919-4	1108	X	X
69-28727-8	1133	X	X	65-69659-1	1142A	X	X
69-36670-1	1134	X	X	65-70950-1	1138	X	
69-37478-1	1135	X	X	65-70950-2	1138		X
69-45172-1	1136	X	X	69-15863-1	1130	X	X
69-48936-1	1137	X	X	69-48936-1	1137	X	X
69-50911-2 *[1]	1141	X	X	JT8D9BG3		X	X
JT8D17ABG314		X	X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (FE)				AIRLINE (FF)			
MODEL 2S2C				MODEL 2T4			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-21950-5	1104	X		65-21950-5	1104	X	
65-21950-6	1104		X	65-21950-6	1104		X
65-22249-2	1105	X	X	65-22249-2	1105	X	X
65-45799-165	1109	X	X	65-45798-5	1109	P	P
65-46851-4	1111	X	X	65-46851-5	1111	P	P
65-52971-1	1117	X	X	65-46851-4	1111	P	P
65-51534-9	1110	X	X	65-51534-5	1110	X	X
65-55475-28	1119	X	X	65-52971-1	1117	X	X
65-58125-3	1124	X	X	65-58125-3	1124	X	X
65-58135-20	1126		X	65-58126-20	1125	P	P
65-58135-18	1126	X		65-58135-21	1126	P	
65-58135-30	1126	X	X	65-58135-30	1126	P	P
65-58145-12	1127	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-1	1142A	X	X	65-65919-4	1108	X	X
65-70950-11	1138	X		65-69659-1	1142A	X	X
65-70950-12	1138		X	65-70950-7	1138	P	
65-85379-3	1143	X		65-70950-8	1138		P
65-85379-4	1143		X	65-70950-11	1138	P	
69-28727-8	1133	X	X	65-70950-12	1139		P
69-48936-1	1137	X	X	65-77451-4	1125	P	P
JT8D17BG301		X	X	65-85379-3	1143	X	
JT8D17BG42 (OPT)		X	X	65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D15BG301*[2]		P	P
				JT8D17BG301*[1]*[3]		P	P
				JT8D17BG300BFE*[3]		P	

*[1] ENGINE INTERMIX WITH JT8D-9, -9A, -15 PER MC 3510-102

*[2] ENGINE INTERMIX WITH JT8D-17 PER MC 3510-124K, AND SB 71-1078

*[3] REPLACE ONE JT8D17BG301 ENGINE WITH ONE JT8D17BG300 PER MC 3510-143

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (FF)
 MODEL 2Q9, 2T4
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (FF)
 MODEL 112
 ENGINE 1 65-73769-45
 ENGINE 2 65-73769-46

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-19684-1	1102	X	X
65-21950-6	1104		X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-2	1105	X	X
65-22454-8	1106	P	P	65-22454-8	1106	X	X
65-45799-1	1109	X	X	65-45798-1	1109	X	X
65-46851-4	1111	X	X	65-46853-1	1113	X	X
65-51534-5	1110	X	X	65-51534-1	1110	X	X
65-58125-3	1124	X	X	65-52261-1	1116	X	X
65-58135-18	1126	X		65-52971-1	1117	X	X
65-58135-20	1126		X	65-55750-1	1121	X	
65-65919-4	1108	X	X	65-55750-2	1121		X
65-69659-1	1142A	X	X	65-57815-1	1140	X	X
65-70950-7	1138	X		65-57815-2	1140	X	X
65-70950-8	1138		X	65-58125-1	1124	X	X
65-85379-3	1143	X		65-58135-18	1126	X	
65-85379-4	1143		X	65-58135-20	1126		X
69-15803-1	1130	X	X	65-58145-4	1127	X	X
69-48936-1	1137	X	X	65-58234-10	1128	X	X
JT8D15BG301		X	X	65-58234-3	1128	X	X
				65-65919-4	1108	X	X
				65-69659-1	1142A	X	X
				65-70950-1	1138	X	
				65-70950-2	1138		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D9BG3 *[1] *[2]		P	P
				JT8D9BG9 *[1] *[2]		P	P
				JT8D-9A *[2]		P	P
				JT8D-7B *[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] ENGINE INTERMIX WITH JT8D-7, -7A, -7B PER MC 3510-122K AND SB 71-1087.
 *[2] ENGINE INTERMIX PER MC 3510-126K AND SB 71-1077

OVERHAUL MANUAL

AIRLINE (FL)				AIRLINE (FL)			
MODEL 222				MODEL 247			
ENGINE 1 65-73769-5				ENGINE 1 65-73769-7			
ENGINE 2 65-73769-6				ENGINE 2 65-73769-8			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-1	1105	X	X	65-22249-3	1105	X	X
65-22454-8	1106	X	X	65-22454-7	1106	X	X
65-25736-3	1107	X	X	65-27808-1	1108	X	X
65-45798-1	1109	X	X	65-45798-1	1109	X	X
65-46853-1	1113	X	X	65-46853-1	1113	X	X
65-50571-2	1114	X	X	65-50882-2	1115	X	X
65-51534-1	1110	X	X	65-51534-1	1110	X	X
65-52261-1	1116	X	X	65-52261-1	1116	X	X
65-52971-1	1117	X	X	65-52971-1	1117	X	X
65-55750-1	1121	X		65-55475-21	1119	X	X
65-55750-2	1121		X	65-55750-1	1121	X	
65-58125-1	1124	X	X	65-55750-2	1121		X
65-58125-2	1124	X	X	65-58125-1	1124	X	X
65-58135-12	1126	X	X	65-58135-18	1126	X	
65-58145-4	1127	X	X	65-58135-20	1126		X
65-58234-9	1128	X	X	65-58145-4	1127	X	X
65-65919-4	1108	X	X	65-58234-10	1128	X	X
65-57815-1	1140	P	P	65-58234-3	1128	X	X
65-57815-2	1140	P	P	65-65919-4	1108	X	X
65-70950-1	1138	P		65-69659-7	1142A	X	X
65-70950-2	1138		P	65-70950-1	1138	X	
65-73778-1 (SB 78-1005)		X		65-70950-2	1138		X
69-15863-1	1130	X	X	69-15863-1	1130	X	X
69-48936-1	1137	X	X	69-21612-1	1132	X	X
JT8D7BG3 *[2]		P	P	69-48936-1	1137	X	X
JT8D9BG2 (SB 71-1043) *[1]		P	P	JT8D9BG4		X	X
JT8D9A *[1]		P	P				
JT8D7 *[2]		P	P				
JT8D7A *[2]		P	P				
JT8D15 *[2]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] ENGINE INTERMIX WITH JT8D7 PER MC 3510-130K AND SB 71-1079
*[2] ENGINE INTERMIX WITH JT8D9 PER MC 3510-123K AND SB 71-1088

OVERHAUL MANUAL

AIRLINE (FL)

MODEL 214, 2H4
 ENGINE 1 65-73769-37
 ENGINE 2 65-73769-38

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-13	1106	P	P
65-22454-8	1106	P	P
65-27808-1	1108	P	P
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1121	X	
65-55750-2	1121		X
65-57815-1	1140	P	P
65-57815-2	1140	P	P
65-58125-1	1124	X	X
65-58135-12	1126	P	P
65-58135-18	1126	P	
65-58135-20	1126		P
65-58145-1	1127	X	X
65-58145-4	1127	X	X
65-58234-10	1128	P	P
65-58234-3	1128	X	X
65-65919-4	1108	P	P
65-69659-2	1142A	P	P
65-69659-10	1142A	P	P
65-70950-1	1138	P	
65-70950-2	1138		P
65-70950-5	1138	P	
65-70950-6	1138		P
65-73778-1	(SB 78-1005)	P	P
69-15863-1	1130	P	P
69-28727-1	1133	P	P
69-48936-1	1137	X	X
JT8D7BG3		P	P
JT8D9BG3		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (FL)

AIRLINE (FL)

MODEL 212, 291, 2A1
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	JT8D17*[2]*[3]		P	P
65-20996-11	1103	P	P	JT8D9*[3]		P	P
65-21950-3	1104	P	P	JT8D9A*[3]		P	P
65-21950-5	1104	P	P				
65-21950-6	1104	P	P				
65-22249-2	1105	X	X				
65-22454-8	1106	P	P				
65-22454-9	1106	P	P				
65-45798-1	1109	P	P				
65-46851-4	1111	P	P				
65-46853-1	1113	P	P				
65-51534-1	1110	P	P				
65-51534-5	1110	P	P				
65-52261-1	1116	P	P				
65-52971-1	1117	P	P				
65-55750-1	1121	P		*[1] ENGINE INTERMIX PER MC 3510-72, 3510-213K, 3510-146K, AND SB 71-1106, 71-1138			
65-55750-2	1121		P				
65-58125-1	1124	P	P				
65-58125-3	1124	P	P				
65-58135-18	1126	X		*[2] ENGINE INTERMIX PER MC 3510-149K AND SB 71-1096			
65-58135-20	1126		X				
65-58135-30	1126	P	P				
65-58145-4	1127	P	P				
65-58234-10	1128	X	X	*[3] ENGINE INTERMIX PER MC 3510-225K AND SB 71-1154, AND MC 3510-232K			
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-70950-1	1138	P					
65-70950-11	1138	P					
65-70950-12	1138		P				
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-85379-3	1143	P					
65-85379-4	1143		P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D17BG301*[1]		P	P				
JT8D17A*[1]*[2]*[3]	P	P					
JT8D9ABG42		P	P				
JT8D9BG9*[1]		P	P				
JT8D9ABG301*[1]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (FL)
MODEL 2C0, 291
ENGINE 1 65-73769-55
ENGINE 2 65-73769-56

AIRLINE (GF)
MODEL 2P6
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19864-1	1102	X	X	65-20996-1*[1]	1103	X	X
65-21950-3	1104	X	X	65-21950-5	1104	X	
65-22249-2	1105	X	X	65-21950-6	1104		X
65-22458-8	1106	X	X	65-22249-3	1105	X	X
65-45798-1	1109	X	X	65-46851-4	1111	X	X
65-46853-1	1113	X	X	65-51534-1	1110	P	P
65-51534-1	1110	X	X	65-51534-5	1110	P	P
65-52261-1	1116	X	X	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-55475-26	1119	X	X
65-55750-1	1121	X		65-58125-3	1124	X	X
65-55750-2	1121		X	65-58135-18	1126	X	
65-57815-1	1140	P	P	65-58135-20	1126		X
65-57815-2	1140	P	P	65-58234-3	1128	X	X
65-58125-1	1124	X	X	65-65919-4	1108	X	X
65-58135-12	1126	P	P	65-70950-7	1138	X	
65-58135-18	1126	P		65-70950-8	1138		X
65-58135-20	1126		P	65-85379-3	1143	X	
65-58145-4	1127	X	X	65-85379-4	1143		X
65-58234-10	1128	P	P	65-69659-1	1142A	X	X
65-58234-3	1128	X	X	69-15863-1	1130	X	X
65-65919-4	1108	X	X	69-48936-1	1137	X	X
65-69659-1	1142A	X	X	69-56334-1	1139	X	X
65-70950-1	1138	P		JT8D15BG300		X	X
65-70950-2	1138		P	JT8D15BG41 (OPT)			X
65-70950-5	1138	P					
65-70950-6	1138		P				
65-73778-1 (SB 78-1005)		P	P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D9BG3		X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
PER SB 80-1004, 80-1030

AIRLINE (GI)				AIRLINE (GN)			
MODEL 2R6C ENGINE 1 65-73769-23 ENGINE 2 65-73769-24				MODEL 2Q2C ENGINE 1 65-73769-23 ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		61-30196-001	1117	X	X
65-21950-6	1104		X	61-30198-125	1117	X	X
65-22249-3	1105	X	X	61-30200-001	1117	X	X
65-45798-5	1109	X	X	61-30404-3	1117	X	X
65-46851-5	1111	X	X	65-21950-5	1104	X	
65-51534-5	1110	X	X	65-21950-6	1104		X
65-52971-1	1117	X	X	65-22249-1	1105	X	X
65-58125-3	1124	X	X	65-22454-8	1106	X	X
65-58126-20	1125	X	X	65-45798-5	1109	X	X
65-58135-18	1126	X		65-46851-5	1111	X	X
65-58135-20	1126		X	65-58125-3	1124	X	X
65-58135-21	1126	X		65-58126-20	1125	X	X
65-58135-30	1126	X	X	65-58135-21	1126	X	
65-65919-4	1108	X	X	65-58135-26	1126	X	X
65-70950-11	1138	X		65-58234-3	1128	X	X
65-70950-12	1138		X	65-65919-4	1108	X	X
65-77451-4	1125	X	X	65-70950-11	1138	X	
65-85379-3	1143	X		65-70950-12	1138		X
65-85379-4	1143		X	65-77451-3	1125	X	X
69-15863-1	1130	X		65-85379-3	1143	X	
69-48936-1	1137	X	X	65-85379-4	1143		X
JT8D17BG301		X	X	69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D17BG301		X	X
				65C21828-70 *[1]		P	P
				JT8D-15 *[1] *[2]		P	P
				JT8D-17 *[1] *[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC 7200MK3051 AND SB 737-71-1312.

*[2] ENGINE INTERMIX PER MC 7200MK3265 AND SB 737-71-1340.

POWER PACK INSTALLATIONS
(GI) (GN)

AIRLINE (GUN)

MODEL 247
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-56646-3	*[1] 1122	X	X
65-19592-24	*[1] 1101		X	65-58122-2	*[1] 1123	X	X
65-20996-21	*[1] 1103	X	X	65-58125-3	1124	X	X
65-21931-23	1114A	X	X	65-58135-18	1126	X	
65-21950-5	1104	X		65-58135-20	1126		X
65-21950-6	1104		X	65-58135-47	*[1] 1126	X	X
65-22249-2	1105	X	X	65-58234-3	1128	X	X
65-45798-4	1109	X	X	65-58234-20	1128	X	X
65-45798-6	1109	X	X	65-65919-4	1108	X	X
65-45798-7	1109	X	X	65-69659-1	1142A	X	X
65-45798-10	1109	X	X	65-70950-7	1138	X	
65-45799-165	1109	X	X	65-70950-8	1138		X
65-45799-170	1109	X	X	65-85379-3	1143	X	
65-46851-4	*[1] 1111	X	X	65-85379-4	1143		X
65-50571-5	*[1] 1114	X	X	69-15863-1	1130	X	X
65-51534-8	1110	X	X	69-50911-2	*[1] 1141	X	X
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	X	X	JT8D15ABG304		X	X
65-55475-33	*[1] 1119	X	X				
65-55749-8	*[1] 1120	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (GUN)
MODEL 2T4
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X					
65-19592-24	*[1] 1101		X	65-58135-47	*[1] 1126	X	X
65-20996-24	*[1] 1103	X	X	65-58145-12	*[1] 1127	X	X
65-21950-5	1104	X		65-58234-3	1128	X	X
65-21950-6	1104		X	65-65919-4	1108	X	X
65-22249-2	1105	X	X	65-69659-1	1142A	X	X
65-25736-2	1107	X	X	65-70950-11	1138	X	
65-45798-3	1109	X	X	65-70950-12	1138		X
65-45798-4	1109	X	X	65-77451-4	1125	X	X
65-45798-5	1109	X	X	65-85379-3	1143	X	
65-45798-10	1109	X	X	65-85379-4	1143		X
65-45799-150	1109	X	X	69-15935-1	1131	X	X
65-46851-5	*[1] 1111	X	X	69-28727-8	1133	X	X
65-46852-1	1112	X	X	69-36670-1	1134	X	X
65-46853-2	1113	X	X	69-37478-1	1135	X	X
65-46858-131	1138	X		69-45172-1	1136	X	X
65-46858-132	1138		X	69-48936-1	1137	X	X
65-48361-	WIRE BUNDLE & TUBING			69-50911-1	1141	X	X
65-50571-1	1114	X	X	JT8D17AB6300		X	X
65-50882-1	1115	X	X				
65-51534-5	1110	X	X				
65-52261-2	1116	X	X				
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	X	X				
65-55475-28	*[1] 1119	X	X				
65-55749-9	*[1] 1120	X	X				
65-56646-1	1122	X	X				
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-20	1125	X	X				
65-58126-24	1125	X	X				
65-58135-1	1126	X	X				
65-58135-3	1126	X	X				
65-58135-20	1126		X				
65-58135-21	1126	X					
65-58135-30	*[1] 1126	X	X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (GN)

MODEL 2Q2C

ENGINE 1 65-73769-23

ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-3 * ^[1]	1101	X	X	65-58234-3 * ^[1]	1128	X	X
65-19684-1 * ^[1]	1102	X		65-65919-4 * ^[1]	1108	X	X
65-20996-1 * ^[1]	1103	X	X	65-70950-5 * ^[1]	1138	X	
65-21950-5 * ^[1]	1104	X		65-70950-6 * ^[1]	1138		X
65-21950-6 * ^[1]	1104		X	65-85379-3 * ^[1]	1143	X	
65-25736-2 * ^[1]	1107	X	X	65-85379-4 * ^[1]	1143		X
65-45798-3 * ^[1]	1109	X	X	69-15935-1 * ^[1]	1131	X	X
65-46851-1 * ^[1]	1111	X	X	69-36670-1 * ^[1]	1134	X	X
65-46852-1 * ^[1]	1112	X	X	69-37478-1 * ^[1]	1135	X	X
65-46853-2 * ^[1]	1113	X	X	69-45172-1 * ^[1]	1136	X	X
65-50571-1 * ^[1]	1114		X	69-48936-1 * ^[1]	1137	X	X
65-50882-1 * ^[1]	1115	X	X	69-50911-1 * ^[1]	1141	X	X
65-52261-2 * ^[1]	1116	X	X	65-22249-1 * ^[1]	1105	X	X
65-54476-1 * ^[1]	1118	X	X	65-22454-8 * ^[1]	1106	X	X
65-55475-1 * ^[1]	1119	X	X	65-45798-5 * ^[1]	1109	X	X
65-55749-1 * ^[1]	1120	X	X	65-46851-5 * ^[1]	1111	X	X
65-56646-1 * ^[1]	1122	X	X	65-52971-1 * ^[1]	1117	X	X
65-58122-1 * ^[1]	1123	X	X	65-58126-20 * ^[1]	1125		X
65-58125-3 * ^[1]	1124	X	X	65-58126-25 * ^[1]	1125	X	
65-58126-1 * ^[1]	1125	X	X	65-58135-21 * ^[1]	1126	X	
65-58126-2 * ^[1]	1125	X	X	65-58135-26 * ^[1]	1126	X	X
65-58135-1 * ^[1]	1126	X	X	65-70950-11	1138	X	
65-58135-12 * ^[1]	1126	X		65-70950-12	1138		X
65-58135-2 * ^[1]	1126	X		65-77451-3	1125	X	X
65-58135-20 * ^[1]	1126		X	69-15863-1	1130	X	X
65-58135-3 * ^[1]	1126	X	X	JT8D17BG301 * ^[2]		X	X
65-58135-4 * ^[1]	1126	X	X	JT8D-15 * ^[2] * ^[3]		X	X
65-58145-10 * ^[1]	1127	X	X	JT8D-15A * ^[3]		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

^[1] SUPERSEDES INSTALLATION LISTED IN TABLE C.^[2] INTERMIX JT8D-17 WITH JT8D-15 ENGINES PER SB 71-1312.*^[3] INTERMIX JT8D-15 WITH JT8D-15A ENGINES PER SB 71-1341.

POWER PACK INSTALLATIONS
(GN)

AIRLINE (HF)				AIRLINE (HM)			
MODEL 2K5				MODEL 2T5			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-31			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-32			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-21950-5	1104	X	X	65-21950-5	1104	X	X
65-21950-6	1104	X	X	65-22249-3	1105	X	X
65-22749-1	1105	X	X	65-46851-4	1111	X	X
65-45798-6	1109	X	X	65-52971-1	1117	X	X
65-45798-8	1109	X	X	65-55475-28	1119	X	X
65-45799-1	1109	X	X	65-58125-3	1124	X	X
65-46851-4	1111	X	X	65-58135-18	1126	X	
65-51534-8	1110	X	X	65-58135-20	1126		X
65-55475-28	1119	X	X	65-58145-12	1127	X	X
65-58125-3	1124	X	X	65-65919-4	1108	X	X
65-58135-18	1126	X		65-70950-5	1138	X	
65-58135-20	1120		X	65-70950-6	1138		X
65-58135-30	1126	X	X	65-70950-7	1138	X	
65-58145-12	1127	X	X	65-70950-8	1138		X
65-65919-4	1108	X	X	65-85379-3	1143	X	
65-69659-1	1142A	X	X	65-85379-4	1143		X
65-70950-11	1138	X		69-21612-1	1132	X	X
65-70950-12	1138		X	69-28727-8	1133	X	X
65-85379-3	1143	X	X	69-48936-1	1137	X	X
69-28727-8	1133	X	X	JT8D15BG301		X	X
69-36670-2	1134	X	X				
69-48936-1	1137	X	X				
69-56334-1	1139	X	X				
JT8D17B300		P	P				
JT8D15 *[1]		P	P				
JT8D17 *[1]		P	P				

*[1] ENGINE INTERMIX PER MC 3510-228K AND SB 71-1163

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

AIRLINE (HV)

MODEL 2K2C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-3	1104	X	X
65-22249-1	1105	X	X
65-22454-8	1106	X	X
65-46851-4	1111	X	X
65-51534-1	1110	X	X
65-52971-1	1117	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D15BG42*[2]		P	P
JT8D15A*[2]		P	P
JT8D15 *[3]		P	P
JT8D17 *[3]*[4]		P	P
JT8D17A *[4]		P	P

*[2] ENGINE INTERMIX PER MC 3510-156K AND SB 71-1107
INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE
ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[3] ENGINE INTERMIX AND REPLACEMENT PER SB 71-1180 AND MC 3510-236K

*[4] ENGINE INTERMIX AND REPLACEMENT PER SB 71-1219 AND MC 7200MK3071

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (HV)

MODEL 222

ENGINE 1 65-73769-5

ENGINE 2 65-73769-6

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	JT8D7BG22		P	P
65-21950-3	1104	X	X	JT8D7BG3		P	P
65-22249-1	1105	X	X	JT8D9BG (3) *[1]		P	P
65-22454-8	1106	X	X	JT8D9ABG (3) *[1]		P	P
65-25736-3	1107	X	X	JT8D7A *[2]		P	P
65-27808-1	1108	P	P	JT8D15 *[2]		P	P
65-45798-1	1109	X	X	JT8D15A *[2]		P	P
65-46851-1	1111	X	X				
65-46853-1	1113	X	X				
65-50571-2	1114	P	P				
65-51534-1	1110	X	X				
65-52261-1	1116	X	X				
65-52971-1	1117	X	X				
65-55750-1	1121	X					
65-55750-2	1121		X				
65-58125-1	1124	X	X				
65-58125-2	1124	P	P				
65-58135-12	1126	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58145-4	1127	X	X				
65-58234-11	1128	P	P				
65-58234-9	1128	X	X				
65-65919-4	1108	P	P				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-73778-1 (SB 78-1005)		P	P				
65-85379-3	1143	P					
65-85379-4	1143		P				
69-15863-1	1130	X	X				
69-48936-1	1137	P	P				

*[1] ENGINE INTERMIX WITH JT8D7 PER MC 3510-35K

*[2] ENGINE INTERMIX PER MC 7200MK3010

AIRLINE (HV)
MODEL 2K2
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (IA)
MODEL 270C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[1]	1103	X	X	65-20996-10	1103	P	P
65-21950-5	1104	X		65-21950-3	1104	P	P
65-21950-6	1104		X	65-21950-5	1104	P	
65-22249-1	1105	X	X	65-21950-6	1104		P
65-46851-4	1111	X	X	65-22249-1	1105	X	X
65-51534-1	1110	X	X	65-22454-8	1106	X	X
65-51534-5	1110	X	X	65-50882-2	1115	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55475-28	1119	X	X	65-52971-1	1117	X	X
65-58125-3	1124	X	X	65-58125-3	1124	X	X
65-58135-18	1126	X		65-58135-18	1126	X	
65-58135-20	1126		X	65-58135-20	1126		X
65-58145-12	1127	X	X	65-58234-10	1128	X	X
65-58234-3	1128	X	X	65-58234-3	1128	X	X
65-65919-4	1108	X	X	65-65919-4	1108	X	X
65-69659-1	1142A	X	X	65-69659-1	1142A	X	X
65-70950-7	1138	X		65-70950-7	1138	X	
65-70950-8	1138		X	65-70950-8	1138		X
65-85379-3	1143	X		65-85379-3	1143	X	
65-85379-4	1143		X	65-85379-4	1143		X
69-28727-8	1133	X	X	69-15863-1	1130	X	X
69-48936-1	1137	X	X	69-48936-1	1137	X	X
JT8D15BG301*[2]*[3]		P	P	JT8D15BG42		X	X
JT8D15BG42 (OPT)*[2]*[3]		P	P				
JT8D15A*[2]		P	P				
JT8D-17*[3]		P	P				
JT8D15*[4]*[5]		P	P				
JT8D17A*[4]		P	P				
JT8D9A*[5]		P	P				

*[1] INCLUDES ENGINE STARTER VALVE OPERATING INDICATOR LIGHT PRESSURE SWITCH

*[2] ENGINE INTERMIX PER MC 3510-156K AND SB 71-1107

*[3] ENGINE INTERMIX PER MC 3510-229K AND SB 71-1155

*[4] ENGINE INTERMIX PER MC 3530-217K AND SB 71-1159

*[5] ENGINE INTERMIX PER MC 3510-230K AND SB 71-1161

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (IC)

MODEL 2A8
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-58234-10	1128	X	X
65-20996-1 *[1]	1103	X	X	65-58234-3	1128	X	X
65-21950-2	1104	P	P	65-65919-4	1108	X	X
65-21950-3	1104	P	P	65-69659-1	1142A	X	X
65-21950-5	1104	P		65-70950-1	1138	P	
65-21950-6	1104		P	65-70950-2	1138		P
65-21950-9	1104	P		65-70950-15	1138	P	
65-21950-10	1104		P	65-70950-16	1138		P
65-22249-1	1105	X	X	65-85379-1	1143	P	
65-22454-8	1106	X	X	65-85379-2	1143		P
65-45798-1	1109	P1	P1	65-85379-3	1143	P	
65-46851-4	1111	P	P	65-85379-4	1143		P
65-46853-1	1113	P1	P1	69-15863-1	1130	P	P
65-51534-1	1110	X	X	69-28727-1	1133	P	P
65-51534-5	1110	X	X	69-48936-1	1137	X	X
65-52261-1	1116	P1	P1	69-56334-1	1139	X	X
65-52971-1	1117	X	X	JT8D9ABG52		P	P
65-55750-1	1120	P		JT8D9BG17		P	
65-55750-2	1120		P	JT8D17BG300 *[2]		P	P
65-58125-1	1124	P	P	JT8D17A *[2]		P	P
65-58125-3	1124	P	P				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-30	1126	P	P				
65-58145-1	1127	P	P				
65-58145-4	1127	P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

P1 = RESTRICTED TO USAGE ON ACOUSTICALLY-TREATED ENGINES

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
 PER MC3540-11K AND SB 80-1029

*[2] ENGINE INTERMIX PER MC 3510-146K AND SB 71-1133

AIRLINE (IC)
MODEL 2A8C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-1 *[1]	1103	X	X
65-21950-9	1104	X	
65-21950-10	1104		X
65-22249-1	1105	X	X
65-22454-8	1106	X	X
65-46851-4	1111	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-30	1126	X	X
65-65919-4	1108	X	X
65-70950-15	1138	X	
65-70950-16	1138		X
65-85379-1	1143	X	
65-85379-2	1143		X
69-48936-1	1137	X	X
69-56334-1	1139	X	X
JT8D17BG300*[2]		P	P
JT8D17A*[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1029

*[2] ENGINE INTERMIX PER MC 3510-146K AND SB 71-1133

AIRLINE (IDF)

MODEL 2A8

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19592-23 *[1]	1101	X	
65-19592-24 *[1]	1101		X
65-20996-1 *[2]	1103	X	X
65-21950-9	1104	X	
65-21950-10	1104		X
65-22249-1	1105	X	X
65-22454-8	1106	X	X
65-45798-10	1109	X	X
65-45799-148	1109	X	X
65-45799-170	1109	X	X
65-46851-4 *[1]	1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-54476-4 *[1]	1118	X	X
65-55749-9 *[1]	1120	X	X
65-58122-2 *[1]	1123	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-30 *[1]	1126	X	X
65-58135-47 *[1]	1126	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
69-56334-1	1139	X	X
JT8D17ABG300		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1029

AIRLINE (IE)
 MODEL 2Q8
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
61-30196-001	1117	X	X
61-30198-170	1117	X	X
61-30200-001	1117	X	X
61-30404-3	1117	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-51534-5	1110	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D9ABG301		X	X
JT8D9ABG42 (OPT)		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (IE)			AIRLINE (IE)		
MODEL 2Q8			MODEL 2Q8		
ENGINE 1 65-73769-51			ENGINE 1 65-73769-51		
ENGINE 2 65-73769-52			ENGINE 2 65-73769-52		
INSTALLATION	FIGURE	ENGINE 1 2	INSTALLATION	FIGURE	ENGINE 1 2
65-19592-23 *[1]	1101	X	65-21950-5	1104	X
65-19592-24 *[1]	1101	X	65-21950-6	1104	X
65-20996-21 *[1]	1103	X X	65-22249-2	1105	X X
65-21950-5	1104	X	65-45799-1	1109	X X
65-21950-6	1104	X	65-46851-4 *[1]	1111	X X
65-22249-2	1105	X X	65-51534-5	1110	X X
65-45798-4	1109	X X	65-52971-1	1117	X X
65-45798-10	1109	X X	65-58125-3	1124	X X
65-45799-1	1109	X X	65-58135-18	1126	X
65-46851-4 *[1]	1111	X X	65-58135-20	1126	X
65-51534-5	1110	X X	65-58135-30 *[2]	1126	X X
65-52971-1	1117	X X	65-58234-3	1128	X X
65-54476-4 *[1]	1118	X X	65-69659-1	1142A	X X
65-55749-9 *[1]	1120	X X	65-70950-11 *[2]	1138	X
65-58122-2 *[1]	1123	X X	65-70950-12 *[2]	1138	X
65-58125-3	1124	X X	65-85379-3	1143	X
65-58135-18	1126	X	65-85379-4	1143	X
65-58135-20	1126	X	69-15863-1	1130	X X
65-58135-47 *[1]	1126	X X	69-48936-1	1137	X X
65-58234-3	1128	X X	JT8D17BG300 *[2]		P P
65-69659-1	1142A	X X	JT8D17 *[3]		P P
65-70950-7	1138	X	JT8D9 *[3]		P P
65-70950-8	1138	X	JT8D9A *[3]		P P
65-85379-3	1143	X			
65-85379-4	1143	X			
69-15863-1	1130	X X			
69-48936-1	1137	X X			
JT8D15ABG300		X X			

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] REPLACEMENT OF JT8D15 ENGINE WITH JT8D17 PER RR 97065-38

*[3] ENGINE INTERMIX PER MC 3510-220K

AIRLINE (IE)

MODEL 2T5
 ENGINE 1 65-73769-31
 ENGINE 2 65-73769-32

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19592-3	1101	P	P
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45799-165	1109	X	X
65-46851-4 *[1]	1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-55475-28 *[1]	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-21612-1	1132	X	X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D15BG301		P	P
JT8D9 *[2]		P	P
JT8D9A *[2]		P	P
JT8D15 *[2]		P	P
JT8D17 *[2]		P	P
JT8D17A *[2]		P	P

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
 *[2] ENGINE INTERMIX PER MC 3510-218K

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (IN)				AIRLINE (IN)			
MODEL 248				MODEL 248C			
ENGINE 1 65-73769-19				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-20				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-20996-10*[1]	1103	X	X
65-20996-1*[1]	1103	X	X	65-21950-3	1104	X	X
65-21950-3	1104	X	X	65-22249-1	1105	X	X
65-22249-1	1105	X	X	65-22454-9	1106	X	X
65-22454-8	1106	P	P	65-46851-4	1111	X	X
65-22454-9	1106	P	P	65-51534-1	1110	X	X
65-45798-1	1109	X	X	65-52971-1	1117	X	X
65-46853-1	1113	X	X	65-55475-28	1119	X	X
65-51534-1	1110	X	X	65-58125-3	1124	X	X
65-51534-5	1110	X	X	65-58135-18	1126	X	
65-52261-1	1116	X	X	65-58135-20	1126		X
65-52971-1	1117	X	X	65-58145-12	1127	X	X
65-55750-1	1120	X		65-58234-10	1128	X	X
65-55750-2	1120		X	65-58234-3	1128	X	X
65-58125-1	1124	X	X	65-65919-4	1108	X	X
65-58135-18	1126	X		65-69659-1	1142A	P	P
65-58135-20	1126		X	65-69659-2	1142A	P	P
65-58145-1	1127	X	X	65-70950-5	1138	X	
65-58145-4	1127	X	X	65-70950-6	1138		X
65-58234-10	1128	X	X	65-85379-3	1143	X	
65-58234-3	1128	X	X	65-85379-4	1143		X
65-65919-4	1108	X	X	69-28727-8	1133	X	X
65-69569-1	1142A	X	X	69-48936-1	1137	X	X
65-70950-1	1138	P		JT8D9ABG55		X	X
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
69-21612-1	1132	X	X				
69-28727-1	1133	X	X				
69-48936-1	1137	X	X				
JT8D9BG10		P	P				
JT8D9BG4		P	P				
JT8D-9A *[2]		P	P				
JT8D-15 *[2]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
PER MC 3540-11K AND SB 80-1025

*[2] ENGINE INTERMIX JT8D-9A WITH JT8D-15 PER SB 71-1258 AND MC 7200MK3032

OVERHAUL MANUAL

AIRLINE (IR)
MODEL 286C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

AIRLINE (IR)
MODEL 286
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-20996-1*[1]	1103	X	X	65-20996-1*[1]	1103	P	P
65-21950-2	1104	X	X	65-20996-10	1103	P	P
65-21950-3	1104	X	X	65-21950-2	1104	X	X
65-22249-2	1105	X	X	65-21950-3	1104	X	X
65-22454-14	1106	P	P	65-22454-20	1106	P	P
65-22454-20	1106	P	P	65-22249-2	1105	X	X
65-45799-134	1109	P	P	65-22454-14	1106	P	P
65-46851-4	1111	P	P	65-45798-1	1109	P	P
65-46853-1	1113	P	P	65-45798-2	1109	P	P
65-51534-1	1110	X	X	65-46851-5	1111	P	P
65-52261-1	1116	P	P	65-46853-1	1113	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55475-28	1119	P	P	65-52261-1	1116	X	X
65-55750-1	1120	X		65-52971-1	1117	X	X
65-55750-2	1120		X	65-55475-28	1119	P	P
65-58125-3	1124	X	X	65-55750-1	1120	X	
65-58135-18	1126	X		65-55750-2	1120		X
65-58135-20	1126		X	65-58125-3	1124	X	X
65-58145-1	1127	P	P	65-58126-20	1125	P	P
65-58145-12	1127	P	P	65-58135-18	1126	P	
65-58145-4	1127	P	P	65-58135-20	1126		P
65-58234-10	1128	X	X	65-58135-21	1126	P	P
65-58234-3	1128	X	X	65-58145-1	1127	P	P
65-65919-4	1108	X	X	65-58145-4	1127	P	P
65-69659-1	1142A	X	X	65-58145-12	1127	P	P
65-70950-1	1138	P		65-58234-10	1128	X	X
65-70950-2	1138		P	65-58234-3	1128	X	X
65-70950-3	1138	P		65-65919-4	1108	X	X
65-70950-4	1138		P	65-69659-1	1142A	X	X
65-70950-7	1138	P		65-70950-1	1138	P	
65-70950-8	1138		P	65-70950-2	1138		P
69-28727-1	1133	P	P	65-70950-7	1138	P	
69-28727-8	1133	P	P	65-70950-8	1138		P
69-48936-1	1137	X	X	65-77451-3	1125	P	P
JT8D15BG42		P	P	69-28727-1	1133	P	P
JT8D9BG21		P	P	69-28727-8	1133	P	P
				69-48936-1	1137	X	X
				JT8D9BG21 1137		P	P
				JT8D15BG42		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
PER MC3540-11K AND SB 80-1014

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (IU)				AIRLINE (IY)			
MODEL 2Q9				MODEL 2N8			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-75			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-76			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-21950-5	1104	X		65-20996-1 *[2]	1103	P	P
65-21950-6	1104		X	65-22249-3	1105	X	X
65-22249-1	1105	P	P	65-22454-24	1106	X	X
65-22249-2	1105	P	P	65-45798-1	1109	X	X
65-22454-8	1106	X	X	65-45798-2	1109	X	X
65-46851-4	1111	P	P	65-46851-5	1111	X	X
65-58125-3	1124	X	X	65-46853-1	1113	X	X
65-58135-18	1126	X		65-51534-1	1110	X	X
65-58135-20	1126		X	65-52261-1	1116	X	X
65-65919-4	1108	X	X	65-52971-1	1117	X	X
65-70950-5	1138	P		65-58126-20	1125	X	X
65-70950-6	1138		P	65-58135-21	1126	X	
65-70950-7	1138	P		65-58145-14	1127	X	X
65-70950-8	1138		P	65-69659-2	1142A	X	X
65-85379-3	1143	X		65-70950-3	1138	X	
65-85379-4	1143		X	65-70950-4	1138		X
69-15863-1	1130	X	X	65-77451-3	1125	X	X
69-48936-1	1137	X	X	65-85379-1	1143	X	
JT8D15BG301 *[1]		P	P	65-85379-2	1143		X
JT8D9ABG301		P	P	65C13071-2	1130	X	X
JT8D7A *[1]		P	P	69-15863-1	1130	X	X
				69-56334-1	1139	X	X
				JT8D15BG48		X	X

*[1] ENGINE INTERMIX WITH JT8D9A PER MC3510-76, MC3510-92

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER SB 80-1039

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (JJ)
 MODEL 2K3
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-58135-1	1126	X	X
65-19592-24	*[1] 1101		X	65-58135-2	1126	X	X
65-20996-24	*[1] 1103	X	X	65-58135-3	1126	X	X
65-21931-23	1114A	X	X	65-58135-18	1126	X	
65-21950-5	1104	X		65-58135-20	1126		X
65-21950-6	1104		X	65-58135-47	*[1] 1126	X	X
65-22249-3	1105	X	X	65-58145-12	*[1] 1127	X	X
65-25736-2	1107	X	X	65-58234-19	1128	X	X
65-45798-3	1109	X	X	65-65919-4	1108	X	X
65-45798-4	1109	X	X	65-69659-10	1142A	X	X
65-45798-7	1109	X	X	65-70950-7	1138	X	
65-45798-10	1109	X	X	65-70950-8	1138		X
65-45799-166	1109	X	X	65-85379-3	1143	X	
65-46851-4	*[1] 1111	X	X	65-85379-4	1143		X
65-46852-1	1112	X	X	69-15935-1	1131	X	X
65-46853-2	1113	X	X	69-28727-8	1133	X	X
65-46868-131	1138	X		69-36670-1	1134	X	X
65-46858-132	1138		X	69-37478-1	1135	X	X
65-50882-1	1115	X	X	69-45172-1	1136	X	X
65-51534-5	1110	X	X	69-48936-1	1137	X	X
65-52261-2	1116	X	X	69-50911-2	*[1] 1141	X	X
65-52971-1	1117	X	X	69-56334-1	1139	X	X
65-54476-4	*[1] 1118	X	X	JT8D15BG314		X	X
65-55475-28	*[1] 1119	X	X	JT8D-15	*[2]	P	P
65-55475-33	*[1] 1119	X	X	JT8D-17	*[2]	P	P
65-55749-8	*[1] 1120	X	X				
65-56646-3	*[1] 1122	X	X				
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-24	1125	X	X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
 *[2] ENGINE INTERMIX JT8D-15 ENGINE WITH JT8D-17 ENGINE PER SB 71-1255 AND MC 7200MK3134

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (KU)
 MODEL 269
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	ENGINE FIGURE	1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-1	1105	X	X
65-22454-9	1106	X	X
65-46851-4	1111	X	X
65-51534-1	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D15BG42		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (LA)				AIRLINE (LA)			
MODEL 2A1				MODEL 2S2C			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-21950-5	1104	X		65-19592-23*[1]	1101	X	
65-21950-6	1104		X	65-19592-24*[1]	1101		X
65-22249-3	1105	X	X	65-21950-5	1104	X	
65-45799-148	1109	X	X	65-21950-6	1104		X
65-46851-4*[1]	1111	X	X	65-22249-2	1105	X	X
65-51534-5	1110	X	X	65-45798-6*[1]	1109	X	X
65-52971-1	1117	X	X	65-45799-148	1109	X	X
65-55475-28*[1]	1119	X	X	65-45799-165	1109	X	X
65-58125-3	1124	X	X	65-46851-4*[1]	1111	X	X
65-58135-18	1126	X		65-51534-9	1110	X	X
65-58135-20	1126		X	65-52971-1	1117	X	X
65-58135-30	1126	X	X	65-55475-28*[1]	1119	X	X
65-58145-12*[1]	1127	X	X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-1	1142A	X	X	65-58135-30	1126	X	X
65-69659-2	1142A	X	X	65-58145-12*[1]	1126	X	X
65-70950-11	1138	X		65-58234-3	1128	X	X
65-70950-12	1138		X	65-65919-4	1108	X	X
65-85379-3	1143	X		65-69659-1	1142A	X	X
65-85379-4	1143		X	65-70950-11	1138	X	
69-28727-8	1133	X	X	65-70950-12	1138		X
69-48936-1	1137	X	X	65-85379-3	1143	X	
JT8D17BG303		P	P	65-85379-4	1143		X
JT8D9 *[2]*[3]		P	P	69-28727-8	1133	X	X
JT8D9A *[2]*[3]		P	P	69-48936-1	1137	X	X
JT8D15 *[2]		P	P	JT8D17BG301		P	P
				JT8D9 *[2]*[3]		P	P
				JT8D9A *[2]*[3]		P	P
				JT8D15 *[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D17 PER MC 3510-102K AND SB 71-1094

*[3] INCLUDES BOTH ACOUSTICALLY TREATED AND UNTREATED ENGINES.

AIRLINE (LG)				AIRLINE (LH)			
MODEL 2C9				MODEL 230			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	P		65-19592-3 *[1]	1101	P	P
65-21950-6	1104		P	65-19592-23 *[1]	1101	P	
65-22249-1	1106	X	X	65-19592-24 *[1]	1101		P
65-22454-9	1106	X	X	65-20996-11 *[1]	1103	X	X
65-46851-4	1111	X	X	65-21931-18	1104A	X	X
65-51534-1	1110	X	X	65-21950-5	1104	X	
65-52971-1	1117	X	X	65-21950-6	1104		X
65-58125-3	1124	X	X	65-22249-3	1105	X	X
65-58135-18	1126	X		65-25736-3	1107	X	X
65-58135-20	1126		X	65-45798-6 *[1]	1109	X	X
65-58234-9	1128	X	X	65-45798-7 *[1]	1109	X	X
65-65919-4	1108	X	X	65-45799-165	1109	X	X
65-69659-1	1142A	X	X	65-45799-166	1109	X	X
65-70950-7	1138	X		65-46851-4 *[1]	1111	X	X
65-70950-8	1138		X	65-51534-8	1110	X	X
65-85379-3	1143	X		65-52971-1	1117	X	X
65-85379-4	1143		X	65-55475-32 *[1]	1119	X	X
69-15863-1	1130	X	X	65-55749-8 *[1]	1120	X	X
69-48936-1	1137	X	X	65-56646-2 *[1]	1122	X	X
JT8D15BG42		X	X	65-58125-3	1124	X	X
				65-58135-18	1126	X	
				65-58135-20	1126		X
				65-58145-12 *[1]	1127	X	X
				65-58234-15	1128	X	X
				65-65919-4	1108	X	X
				65-69659-1	1142A	X	X
				65-69659-2	1142A	P	P
				65-70950-23	1138	X	
				65-70950-24	1138		X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-21612-1	1132	X	X
				69-28727-8	1133	X	X
				69-48936-1	1137	X	X
				69-50911-2 *[1]	1141	X	X
				JT8D17BG309		P	P
				JT8D-17 *[2]		P	P
				JT8D-17 *[2]		P	P
				JT8D-9A *[3]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C.

*[2] REPLACE JT8D-15 ENGINES WITH JT8D-17A, AND INTERMIX ENGINES JT8D-15 OR -17 WITH JT8D-17A PER MC 3510-153K AND SB 71-1099.

*[3] INTERMIX JT8D-15 WITH JT8D-9A ENGINES PER SB 71-1357.

POWER PACK INSTALLATIONS
 (LG) (LH)

AIRLINE (LH)				AIRLINE (LH)			
MODEL 130				MODEL 230C			
ENGINE 1 65-73769-3				ENGINE 1 65-73769-3			
ENGINE 2 65-73769-4				ENGINE 2 65-73769-4			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-1	1105	X	X	65-22249-1	1105	X	X
65-22454-9	1106	X	X	65-22454-9	1106	X	X
65-25736-3	1107	X	X	65-25736-3	1107	X	X
65-27808-1	1108	P	P	65-45798-1	1109	X	X
65-45798-1	1109	X	X	65-46853-1	1113	X	X
65-46853-1	1113	X	X	65-51534-1	1110	X	X
65-51534-1	1110	X	X	65-52261-1	1116	X	X
65-52261-1	1116	X	X	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-55750-1	1120	X	
65-55750-1	1120	X		65-55750-2	1120		X
65-55750-2	1120		X	65-58125-1	1124	X	X
65-58125-1	1124	X	X	65-58135-18	1126	X	
65-58135-12	1126	X	X	65-58135-20	1126		X
65-58145-1	1127	X	X	65-58145-1	1127	X	X
65-58145-4	1127	X	X	65-58145-4	1127	X	X
65-58234-3	1128	X	X	65-58234-10	1128	X	X
65-65919-4	1108	P	P	65-58234-3	1128	X	X
65-69659-2	1142A	P	P	65-65919-4	1108	X	X
65-69659-4	1142A	P	P	65-69659-2	1142A	X	X
65-73778-1	(SB 78-1005)	X	X	65-70950-1	1138	X	
69-21612-1	1132	X	X	65-70950-2	1138		X
69-28727-1	1133	X	X	69-21612-1	1132	X	X
69-48936-1	1137	X	X	69-28727-1	1133	X	X
JT8D7BG3		X	X	69-48936-1	1137	X	X
				JT8D9 *[1]		P	P
				JT8D9A *[1]		P	P
				JT8D9BG3		P	P
				JT8D15 *[1]		P	P

*[1] ENGINE INTERMIX PER MC 3510-208K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737
POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (LT)				AIRLINE (LU)			
MODEL 2A6				MODEL 2Q5C			
ENGINE 1 65-73769-69				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-70				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-19684-1	1102	X	X	61-30196-001	1117	X	X
65-21950-3	1104	X	X	61-30198-195	1117	X	X
65-22249-2	1105	X	X	61-30200-001	1117	X	X
65-22454-8	1106	X	X	61-30404-3	1117	X	X
65-45798-1	1109	X	X	65-21950-5	1104	X	
65-46853-1	1113	X	X	65-21950-6	1104		X
65-51534-1	1110	X	X	65-22249-2	1105	X	X
65-52261-1	1116	X	X	65-45798-2	1109	X	X
65-52971-1	1117	X	X	65-46851-5	1111	X	X
65-55750-1	1120	X		65-50882-1	1115	X	X
65-55750-2	1120		X	65-58125-3	1124	X	X
65-58125-1	1124	X	X	65-58126-20	1125	X	X
65-58135-18	1126	X		65-58135-21	1126	X	
65-58135-20	1126		X	65-58234-3	1128	X	X
65-58145-4	1127	X	X	65-65919-4	1108	X	X
65-58234-10	1128	X	X	65-70950-7	1138	X	
65-58234-3	1128	X	X	65-70950-8	1138		X
65-65919-4	1108	X	X	65-77451-3	1125	X	X
65-69659-7	1142A	X	X	65-85379-3	1143	X	
65-70950-1	1138	X		65-85379-4	1143		X
65-70950-2	1138		X	69-15863-1	1130	X	X
69-15863-1	1130	X	X	69-48936-1	1137	X	X
69-48936-1	1137	X	X	69-56334-1	1139	X	X
JT8D9BG3	X	X		JT8D15BG300		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (LY)
MODEL 258
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23*[1]	1101	X		65-58234-17	1128	X	X
65-19592-24*[1]	1101		X	65-65919-4	1104	X	X
65-20996-19*[1]	1103	X	X	65-69659-1	1142A	X	X
65-21931-18	1104A	X	X	65-70950-11	1138	X	
65-21950-5	1104	X		65-70950-12	1138		X
65-21950-6	1104		X	65-85379-3	1143	X	
65-22249-3	1105	X	X	65-85379-4	1143		X
65-45798-3	1109	X	X	69-15863-1	1130	X	X
65-45798-4	1109	X	X	69-36670-2*[1]	1134	X	X
65-45798-7	1109	X	X	69-48936-1	1137	X	X
65-45798-10	1109	X	X	69-50911-2*[1]	1141	X	X
65-45799-148	1109	X	X	69-56334-1	1139	X	X
65-45799-166	1109	X	X	JT8D17ABG300*[2]		X	X
65-45799-170	1109	X	X				
65-46851-4*[1]	1111	X	X				
65-50571-5*[1]	1114	X	X				
65-51534-5	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4*[1]	1118	X	X				
65-55475-33*[1]	1119	X	X				
65-55749-8*[1]	1120	X	X				
65-56646-2*[1]	1122	X	X				
65-58122-2*[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-30	1126	X	X				
65-58135-47*[1]	1126	X	X				
65-58135-10	1127	X	X				
65-58234-3	1128	X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D-9,-9A,-15,-17 PER CR 05-029 AND 65C25674-8
INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

POWER PACK INSTALLATIONS

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

AIRLINE (MD)				AIRLINE (MD)			
MODEL 2B2				MODEL 2B2			
ENGINE 1 65-73769-33				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-34				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-1	1105	X	X
65-22454-14	1106	X	X	65-22454-20	1106	X	X
65-45798-1	1109	X	X	65-45798-1	1109	X	X
65-46851-4 *[1]	1111	X	X	65-46851-4	1111	X	X
65-46853-1	1113	X	X	65-46853-1	1113	X	X
65-69659-1	1142A	X	X	65-51534-1	1110	X	X
65-52261-1	1116	X	X	65-52261-1	1116	X	X
65-52971-1	1117	X	X	65-52971-1	1117	X	X
65-55750-1	1120	X		65-55750-1	1120	X	
65-55750-2	1120		X	65-55750-2	1120		X
65-58125-1	1124	X	X	65-58125-3	1124	X	X
65-58135-18	1126	X		65-58135-18	1126	X	
65-58135-20	1126		X	65-58135-20	1126		X
65-58145-1	1127	X	X	65-58145-1	1127	X	X
65-58145-4	1127	X	X	65-58145-4	1127	X	X
65-58234-10	1128	X	X	65-58234-10	1128	X	X
65-58234-3	1128	X	X	65-58234-3	1128	X	X
65-65919-4	1108	X	X	65-65919-4	1108	X	X
65-51534-1	1110	X	X	65-69659-1	1142A	X	X
65-70950-1	1138	X		65-70950-3	1138	X	
65-70950-2	1138		X	65-70950-4	1138		X
65-70950-3 *[1]	1138	X		69-28727-1	1133	X	X
65-70950-4 *[1]	1138		X	69-48936-1	1137	X	X
65-70950-5	1138	X		JT8D15BG21		X	X
65-70950-6	1138		X				
69-28727-1	1133	X	X				
69-48936-1	1137	X	X				
JT8D9BG9 ()		X	X				
JT8D15BG ()							

*[1] USED ON JT8D15BG() ONLY MC 3510-14K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 POWER PACK INSTALLATIONS

AIRLINE (EL)				AIRLINE (ET)			
MODEL 2S5C				MODEL 260			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-20996-1*[1]	1103	X	X
65-20996-1*[1]	1103	X	X	65-21950-5	1104	X	
65-21950-3	1104	X	X	65-21950-6	1104		X
65-22249-2	1105	X	X	65-22249-2	1105	X	X
65-22454-9	1106	X	X	65-22454-9	1105	X	X
65-45798-8	1109	P	P	65-46851-4	1111	X	X
65-45799-134	1109	P	P	65-51534-1	1110	X	X
65-46851-4	1111	X	X	65-52971-1	1117	X	X
65-50571-7 *[1]	1114	P	P	65-58125-3	1124	X	X
65-51534-1	1110	X	X	65-58135-18	1126	X	
65-51534-5	1110	X	X	65-58135-20	1126		X
65-52971-1	1117	X	X	65-58234-10	1128	X	X
65-55475-26	1119	P	P	65-58234-3	1128	X	X
65-55750-1	1121	P		65-65919-4	1108	X	X
65-55750-2	1121		P	65-69659-1	1142A	X	X
65-58125-3	1124	X	X	65-70950-3	1138	X	
65-58135-18	1126	X		65-70950-4	1138		X
65-58135-20	1126		X	65-85379-1	1143	X	
65-58234-10	1128	X	X	65-85379-2	1143		X
65-58234-16	1128	P	P	65-85379-3	1143	X	
65-58234-3	1128	P	P	65-85379-4	1143		X
65-65919-4	1108	X	X	69-15863-1	1130	X	X
65-69659-1	1142A	X	X	69-48936-1	1137	X	X
65-70950-3	1138	X		69-56334-1	1139	X	X
65-70950-4	1138		X	JT8D15BG41*[2]		P	P
65-85379-1	1143	P		JT8D15A*[2]		P	P
65-85379-2	1143		P				
65-85379-3	1143	P					
65-85379-4	1143		P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
69-56334-1	1139	X	X				
JT8D15BG31*[2]		P	P				
JT8D15BG41*[2]		P	P				
JT8D15A*[2]		P	P				
JT8D15ABG300		P	P				
JT8D15ABG314		P	P				

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC3540-11K AND SB 80-1017

*[2] ENGINE INTERMIX PER MC 3510-156K AND SB 71-1107
INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (MON)

MODEL 2T7

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	ENGINE FIGURE	1	2
65-20996-1*[1]	1103	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D9ABG42		X	X

*[1] INCLUDE ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1011

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (MI)

Jun 5/88

71-03-01
 Page 1243

AIRLINE (MJ)

MODEL 25A

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-58135-1	1126	X	X
65-19592-24 *[1]	1101		X	65-58135-3	1126	X	X
65-20996-24 *[1]	1103	X	X	65-58135-18	1126	X	
65-21931-21	1114A	X	X	65-58135-20	1126		X
65-21950-5	1104	X		65-58135-30 *[1]	1126	X	X
65-21950-6	1104		X	65-58135-47 *[1]	1126	X	X
65-22249-2	1105	X	X	65-58145-12 *[1]	1127	X	X
65-25736-2	1107	X	X	65-58234-10	1128	X	X
65-45798-3	1109	X	X	65-58234-19	1128	X	X
65-45798-4	1109	X	X	65-65919-4	1108	X	X
65-45798-7	1109	X	X	65-69659-1	1142A	X	X
65-45798-10	1109	X	X	65-70950-11	1138	X	
65-45799-1	1109	X	X	65-70950-12	1138		X
65-45799-148	1109	X	X	65-85379-3	1143	X	
65-45799-166	1109	X	X	65-85379-4	1143		X
65-46851-4 *[1]	1111	X	X	69-15935-1	1131	X	X
65-46852-1	1112	X	X	69-28727-8	1133	X	X
65-46853-2	1113	X	X	69-36670-1	1134	X	X
65-46858-131	1118	X		69-37478-1	1135	X	X
65-46858-132	1118		X	69-45172-1	1136	X	X
65-50882-1	1115	X	X	69-48936-1	1137	X	X
65-51534-5	1110	X	X	69-50911-2 *[1]	1141	X	X
65-52261-2	1116	X	X	JT8D17ABG300		X	X
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X				
65-54475-33 *[1]	1119	X	X				
65-54475-34 *[1]	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-3 *[1]	1122	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-24	1125	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC3510-88

AIRLINE (MON)

MODEL 2T7
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45799-148	1109	X	X
65-46851-4 *[1]	1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-30 *[1]	1126	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17BG301 *[2]*[3]		P	P
JT8D9 *[2]*[3]		P	P
JT8D9A *[2]*[3]		P	P
JT8D15 *[3]		P	P

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER MC 3510-144 AND SB 71-1119

*[3] ENGINE INTERMIX PER MC 3510-201K AND SB 71-1119

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (MR)
 MODEL 2S2C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-1 *[1]	1103	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-45798-6	1109	X	X
65-45799-148	1109	X	X
65-45799-165	1109	X	X
65-46851-4	1111	X	X
65-52971-1	1117	X	X
65-51534-9	1110	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58135-20	1126		X
65-58135-18	1126	X	
65-58135-30	1126	X	X
65-58145-12	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D17BG301		X	X

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1008

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (MS)
 MODEL 266, 2N7
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (MT)
 MODEL 2H5
 ENGINE 1 65-73769-13
 ENGINE 2 65-73769-14

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[1]	1103	P	P	65-19684-1	1102	X	X
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-2	1105	X	X
65-22249-1	1105	X	X	65-22454-9	1106	X	X
65-45799-148	1109	X	X	65-45798-1	1109	X	X
65-46851-4	1111	X	X	65-46853-1	1113	X	X
65-51534-1	1110	X	X	65-51534-1	1110	X	X
65-52971-1	1117	X	X	65-52261-1	1116	X	X
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58135-18	1126	X		65-55750-1	1120	X	
65-58135-20	1126		X	65-55750-2	1120		X
65-58135-26	1126	X	X	65-58125-1	1124	X	X
65-58234-10	1128	X	X	65-58135-18	1126	X	
65-58234-3	1128	X	X	65-58135-20	1126		X
65-65919-4	1108	X	X	65-58145-4	1127	X	X
65-69659-2	1142A	X	X	65-58234-10	1128	X	X
65-70950-11	1138	X		65-65919-4	1108	X	X
65-70950-12	1138		X	65-69659-2	1142A	X	X
65-85379-3	1143	X		65-70950-1	1138	X	
65-85379-4	1143		X	65-70950-2	1138		X
69-15863-1	1130	X	X	69-15863-1	1130	X	X
69-48936-1	1137	X	X	69-48936-1	1137	X	X
JT8D9ABG42		X	X	JT8D7BG3		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] INCLUDE ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1009

AIRLINE (ND)
MODEL 242C
ENGINE 1 65-73769-43
ENGINE 2 65-73769-44

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-73778-1	(SB 78-1005)	P	P
65-21950-3	1104	X	X	65-77451-1	1125	P	P
65-22249-2	1105	X	X	65-77451-2	1125	P	P
65-22454-8	1106	P	P	69-15863-1	1130	X	X
65-22454-9	1106	P	P	69-48936-1	1137	X	X
65-45798-1	1109	X	X	JT8D9BG10		P	P
65-45798-2	1109	X	X	JT8D9BG4		P	P
65-45799-117	1109	P	P				
65-45799-120	1109	P	P				
65-45799-127	1109	P	P				
65-46851-2	1111	P	P				
65-46851-3	1111	P	P				
65-46853-1	1113	X	X				
65-51534-1	1110	X	X				
65-52261-1	1116	X	X				
65-52971-1	1117	X	X				
65-55750-1	1120	X					
65-55750-2	1120		X				
65-58125-1	1124	X	X				
65-58126-14	1125	P	P				
65-58126-20	1125	P	P				
65-58135-12	1126	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58135-21	1126	P					
65-58135-22	1126	P	P				
65-58145-4	1127	X	X				
65-58234-10	1128	P	P				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-2	1142A	X	X				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(ND)

737

POWERPLANT BUILDUP

BOEING

COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (ND)
MODEL 242, 212
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (ND)
MODEL 242C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-21950-5	1104	X	
65-20996-10	1103	P	P	65-21950-6	1104		X
65-21950-2	1104	P	P	65-22249-2	1105	X	X
65-21950-3	1104	P	P	65-22454-8	1106	X	X
65-21950-5	1104	P		65-45798-2	1109	X	X
65-21950-6	1104		P	65-51534	1110	X	X
65-22249-2	1105	P	P	65-52971-1	1117	X	X
65-22454-8	1106	X	X	65-58125-3	1124	X	X
65-45798-1	1109	P	P	65-58126-20	1125	X	X
65-45798-2	1109	P	P	65-58135-18	1126	X	
65-45799-127	1109	P	P	65-58135-20	1126		X
65-46851-3	1111	P	P	65-58135-21	1126	X	
65-46853-1	1113	P	P	65-65919-4	1108	X	X
65-51534-1	1110	P	P	65-69659-1	1142A	X	X
65-51534-5	1110	P	P	65-70950-5	1138	X	
65-52261-1	1116	P	P	65-70950-6	1138		X
65-52971-1	1117	X	X	65-77451-4	1125	X	X
65-55750-1	1120	P		65-85379-3	1143	X	
65-55750-2	1120		P	65-85379-4	1143		X
65-58125-3	1124	X	X	69-15863-1	1130	X	X
65-58126-20	1125	P	P	69-48936-1	1137	X	X
65-58135-18	1126	X		JT8D9ABG306		X	X
65-58135-20	1126		X				
65-58135-21	1126	P					
65-58145-4	1127	P	P				
65-58234-10	1128	X	X				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-77451-3	1125	P	P				
65-85379-3	1143	P					
65-85379-4	1143		P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D9ABG55		P	P				
JT8D9BG21		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(ND)

Jan 5/81

71-03-01
Page 1247

AIRLINE (NE)

MODEL 2W8
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45798-6	1109	X	X
65-45799-148	1109	X	X
65-45799-165	1109	X	X
65-46851-4*[1]	1111	X	X
65-51534-9	1110	X	X
65-52971-1	1117	X	X
65-55475-28*[1]	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-30	1126	X	X
65-58145-12*[1]	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-36670-1	1134	X	X
69-48936-1	1137	X	X
69-56334-1	1139	X	X
JT8D17BG300		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE
 *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (NH)

MODEL 281
 ENGINE 1 65-73769-67
 ENGINE 2 65-73769-68

INSTALLATION	ENGINE FIGURE	ENGINE	
		1	2
65-19864-1	1102	X	X
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-16	1106	X	X
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-51534-1	1110	P	P
65-51534-5	1110	P	P
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1120	X	
65-55750-2	1120		X
65-58125-3	1124	P	P
65-58125-18	1126	X	
65-58135-20	1126		X
65-58145-4	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	P	P
65-69659-2	1142A	P	P
65-70950-1	1138	P	
65-70950-2	1138		P
69-70950-11	1138	P	
69-70950-12	1138		P
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D7BG3		P	P
JT8D7BG9		P	P
JT8D9BG21		P	P
JT8D9BG3		P	P
JT8D9BG9		P	P
JT8D17	(SB 71-1041)	P	P
JT8D15	(SB 71-1363)	P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (NH)

AIRLINE (NU)

MODEL 203

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
61-30196-001	1117	X	X	65-58145-12	1127	X	X
61-30198-170	1117	X	X	65-58234-3	1128	X	X
61-30200-001	1117	X	X	65-65919-4	1108	X	X
61-30404-3	1117	X	X	65-69659-1	1142A	X	X
65-19592-3	1101	P	P	65-70950-11	1138	X	
65-19592-23	1101	P		65-70950-12	1138		X
65-19592-24	1101		P	65-85379-3	1143	X	
65-20996-1 *[1]	1103	P	P	65-85379-4	1143		X
65-20996-19	1103	P	P	69-28727-8	1133	X	X
65-20996-23 *[2]	1103	P	P	69-48936-1	1137	X	X
65-20996-24 *[2]	1103	P	P	JT8D17BG301		P	P
65-21950-5	1104	X		JT8D17BG316		P	P
65-21950-6	1104		X				
65-22249-2	1105	X	X				
65-45798-4	1109	X	X				
65-45798-10	1109	X	X				
65-46851-4	1111	X	X				
65-46852-2	1112	X	X				
65-50571-7 *[1]	1114	P	P				
65-51534-1	1110	P	P				
65-51534-5	1110	P	P				
65-54476-3 *[1]	1118	P	P				
65-54476-4 *[1]	1118	X	X				
65-55475-28	1119	X	X				
65-55749-9 *[1]	1120	P	P				
65-58122-2 *[1]	1123	P	P				
65-58125-3	1124	X	X				
65-58126-24	1125	P	P				
65-58135-26 *[1]	1126	P	P				
65-58135-30 *[1]	1126	P	P				
65-58135-47 *[1]	1126	P	P				

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1019

*[2] INCLUDES ENGINE STARTER VALVE PRESSURE SWITCH RELOCATION PER SB 80-1045 AND 80-1045R1

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (NV)
 MODEL 130
 ENGINE 1 65-73769-3
 ENGINE 2 65-73769-4

AIRLINE (NZ)
 MODEL 219C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19864-1	1102	X	X	65-19592-23*[1]	1101	X	
65-21950-3	1104	X	X	65-19592-24*[1]	1101		X
65-22249-1	1105	X	X	65-20996-11*[1]	1103	X	X
65-22454-8	1106	X	X	65-20996-19*[1]	1103	X	X
65-27808-1	1108	X	X	65-21950-5	1104	X	
65-45798-1	1109	X	X	65-21950-6	1104		X
65-46853-1	1113	X	X	65-22249-3	1105	X	X
65-51534-1	1110	X	X	65-45798-3	1109	X	X
65-52261-1	1116	X	X	65-45798-4	1109	X	X
65-52971-1	1117	X	X	65-45798-10	1109	X	X
65-55750-1	1120	X		65-45799-1	1109	X	X
65-55750-2	1120		X	65-46851-4*[1]	1111	X	X
65-57815-2	1140	X	X	65-51534-5	1110	X	X
65-57815-2	1140	X	X	65-52971-1	1117	X	X
65-58125-1	1124	X	X	65-54476-4*[1]	1118	X	X
65-58135-12	1126	X	X	65-55749-9*[1]	1120	X	X
65-58145-1	1127	X	X	65-58122-2*[1]	1123	X	X
65-58145-4	1127	X	X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-69659-1	1142A	X	X	65-58135-20	1126		X
65-73778-1	(SB 78-1005)	X	X	65-58135-47*[1]	1126	X	X
69-21612-1	1132	X	X	65-58234-3	1128	X	X
69-28721-1	1133	X	X	65-65919-4	1108	X	X
JT8D9BG3		X	X	65-69659-1	1142A	X	X
				65-70950-7	1138	X	
				65-70950-8	1138		X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D15ABG301		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

OVERHAUL MANUAL

AIRLINE (NZ)
MODEL 222
ENGINE 1 65-73769-5
ENGINE 2 65-73769-6

AIRLINE (NZ)
MODEL 219
ENGINE 1 65-73769-35
ENGINE 2 65-73769-36

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-1	1105	X	X	65-22249-3	1105	X	X
65-22454-8	1106	X	X	65-22454-8	1106	X	X
65-25736-3	1107	X	X	65-27808-1	1108	X	X
65-45798-1	1109	X	X	65-46853-1	1113	X	X
65-45799-134	1109	X	X	65-52261-1	1116	X	X
65-46853-1	1113	X	X	65-52971-1	1117	X	X
65-52261-1	1116	X	X	65-55475-26		X	X
65-52971-1	1117	X	X	65-55750-1	1121	X	
65-55475-26		X	X	65-55750-2	1121		X
65-55750-1	1121	X		65-58125-1	1124	X	X
65-55750-2	1121		X	65-58135-12	1126	X	X
65-58125-1	1124	X	X	65-58145-4	1127	X	X
65-58135-12	1126	X	X	65-58234-3	1128	X	X
65-58145-4	1127	X	X	65-73778-1	(SB 78-1005)	X	X
65-58234-9	1128	X	X	69-15863-1	1130	X	X
65-73778-1	(SB 78-1005)	X	X	69-48936-1	1137	X	X
69-15863-1	1130	X	X	JT8D7BG3 *[1]		P	P
69-48936-1	1137	X	X	JT8D9 *[1]		P	P
JT8D7BG3 *[1]		P	P	JT8D9A *[1]		P	P
JT8D9 *[1]		P	P				
JT8D9A *[1]		P	P				

*[1] ENGINE INTERMIX WITH JT8D-9 PER MC 3510-69K & SB 71-1091

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (NZ)
 MODEL 214
 ENGINE 1 65-73769-37
 ENGINE 2 65-73769-38

AIRLINE (NZ)
 MODEL 219
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	P		65-70950-3	1138	P	
65-19592-24	*[1] 1101		P	65-70950-4	1138		P
65-20996-11	*[1] 1103	P	P	65-70950-7	1138	P	
65-20996-23	*[1] 1103	X	X	65-70950-8	1138		P
65-20996-24	*[1] 1103	P	P	65-85379-3	1143	X	
65-20996-25	*[1] 1103	X	X	65-85379-4	1143		X
65-21931-21	1114A	P	P	65C13071-1	1130	X	X
65-21950-5	1104	X		69-15863-1	1130	X	X
65-21950-6	1104		X	69-48936-1	1137	P	P
65-45798-7	1109	P	P	69-50911-2	*[1] 1141	P	P
65-45798-10	1109	P	P	JT8D15BG42	*[2]	P	P
65-45799-1	1109	P	P	JT8D15ABG314		P	P
65-45799-165	1109	P	P	JT8D15A	*[3] *[4]	P	P
65-46851-4	*[1] 1111	X	X	JT8D15	*[3] *[4]	P	P
65-50571-5	*[1] 1114	P	P	JT8D17A	*[4]	P	P
65-51534-1	1110	P	P	JT8D17	*[4]	P	P
65-51534-5	1110	P	P				
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	P	P				
65-55475-33	*[1] 1119	P	P				
65-55749-8	*[1] 1120	P	P				
65-56646-3	*[1] 1122	P	P				
65-58122-2	*[1] 1123	P	P				
65-58125-3	1124	X	X				
65-58126-24	1125	X	X				
65-58126-25	1125	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58135-47	*[1] 1126	P	P				
65-58145-14	*[1] 1127	P	P				
65-58234-3	1128	P	P				
65-58234-19	1128	P	P				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

- *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
- *[2] ENGINE INTERMIX WITH JT8D-7, -7A PER 65C15668-47 & 65C19684-1, -3
- *[3] ENGINE INTERMIX PER MC 3510-226K and SB 71-1174
- *[4] ENGINE INTERMIX PER MC 7200MK3158 AND SB 71-1263

POWER PACK INSTALLATION
 (NZ)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (NZ)

MODEL 214
 ENGINE 1 65-73769-37
 ENGINE 2 65-73769-38

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-13	1106	X	X
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55475-26	1119	X	X
65-55750-1	1121	X	
65-55750-2	1121		X
65-58125-1	1124		
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-4	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-70950-1	1138	X	
65-70950-2	1138		X
65-70950-5	1138	X	
65-70950-6	1138		X
69-28727-1	1133	X	X
69-48936-1	1137	X	X
JT8D7BG3		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (NZ)
MODEL 219
ENGINE 1 65-73769-61
ENGINE 2 65-73769-62

AIRLINE (OA)
MODEL 284
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X				
65-21950-3	1104	X	X	65-20996-1*[1]	1103	X	X
65-22249-2	1105	X	X	65-21950-5	1104	X	
65-22454-9	1106	X	X	65-21950-6	1104		X
65-45798-1	1109	X	X	65-22249-3	1105	X	X
65-46853-1	1113	X	X	65-22454-9	1106	X	X
65-52261-1	1116	X	X	65-45799-1	1109	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55475-26	1119	X	X	65-52971-1	1117	X	X
65-55750-1	1121	X		65-58125-3	1124	X	X
65-55750-2	1121		X	65-58135-18	1126	X	
65-58125-1	1124			65-58135-20	1126		X
65-58135-18	1126	X		65-58234-9	1108	X	X
65-58135-20	1126		X	65-65919-4	1108	X	X
65-58234-3	1128	X	X	65-69659-1	1142A	X	X
65-65919-4	1108	X	X	65-70950-5	1138	X	
65-70950-5	1138	X		65-70950-6	1138		X
65-70950-6	1138		X	65-85379-3	1143	X	
69-15863-1	1130	X	X	65-85379-4	1143		X
69-48936-1	1137	X	X	69-15863-1	1130	X	X
JT8D9BG9		X	X	69-48936-1	1137	X	X
				JT8D9ABG42		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

*[1] INCLUDE ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER
MC 3540-11K AND SB 80-1028

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (OA)
 MODEL 284
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-51

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-1	1105	X	X
65-22454-8	1106	X	X
65-45799-1	1109	X	X
65-51534-1	1110	P	P
65-51534-5	1110	P	P
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-9	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D9ABG301		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

MODEL 2T5 AIRLINE (OI)				MODEL 2UF AIRLINE (OJ)			
ENGINE 1 65-73769-31		ENGINE 2 65-73769-32		ENGINE 1 65-73769-31		ENGINE 2 65-73769-32	
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-19592-3	1101	P	P	65-21950-5	1104	X	
65-19592-23*[1]	1101	P		65-21950-6	1104		X
65-19592-24*[1]	1101		P	65-22249-1	1105	X	X
65-21950-5	1104	X		69-22454-8	1106	X	X
65-21950-6	1104		X	65-45799-1	1109	X	X
65-22249-3	1105	X	X	65-46851-4	1111	X	X
65-45798-10	1109	P	P	65-51534-5	1110	X	X
65-45799-165	1109	X	X	65-55475-28	1119	X	X
65-46851-4*[1]	1111	X	X	65-58125-3	1124	X	X
65-51534-5	1110	X	X	65-58135-18	1126	X	
65-52971-1	1117	X	X	65-58135-20	1126		X
65-54476-4*[1]	1118	P	P	65-58145-12	1127	X	X
65-55475-28*[1]	1119	X	X	65-65919-4	1108	X	X
65-55749-9*[1]	1120	P	P	65-69659-1	1142A	P	P
65-58122-2*[1]	1123	P	P	65-62659-2	1142A	P	P
65-58125-3	1124	X	X	65-70950-7	1138	X	
65-58135-18	1126	X		65-70950-8	1138		X
65-58135-20	1126		X	65-85379-3	1143	X	
65-58135-47	1126	P	P	65-85379-4	1143		X
65-58145-12	1127	X	X	65-28727-8	1133	X	X
65-58234-3	1128	X	X	69-48936-1	1136	X	X
65-65919-4	1108	X	X	JT8D15BG301		X	X
65-69659-1	1142A	P	P				
65-69659-2	1142A	P	P				
65-70950-7	1138	X					
65-70950-8	1138		X				
65-85379-3	1143	X					
65-85379-4	1143		X				
69-21612-1	1132	X	X				
69-28727-8	1133	X	X				
69-48936-1	1137	X	X				
JT8D15BG301		P	P				
JT8D15ABG301		P	P				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

POWER PACK INSTALLATIONS

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (ON)
 MODEL 2L7C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-3	1104	X	X
65-22249-1	1105	X	X
65-22454-21	1106	X	X
65-45798-5	1109	X	X
65-45799-147	1109	P	P
65-45799-148	1109	P	P
65-46851-5	1111	X	X
65-51534-1	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58126-20	1125	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-21	1126	X	
65-58135-26	1126	X	X
65-58234-10	1128	X	X
65-58234-3	1123	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-77451-3	1125	X	X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17BG42		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (ON)

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (PC)				AIRLINE (PH)			
MODEL 2X2				MODEL 2U9			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-20996-1 *[2]	1103	X	X	65-21950-5	1104	X	
65-21950-5	1104	X		65-21950-6	1104		X
65-21950-6	1104		X	65-27749-1	1105	X	X
65-22249-3	1105	X	X	65-45799-148	1109	X	X
65-45798-5	1109	X	X	65-46851-4	1111	X	X
65-46851-5	1111	X	X	65-51534-5	1110	X	X
65-51534-5	1110	X	X	65-58125-3	1124	X	X
65-52971-1	1117	X	X	65-58135-18	1126	X	
65-55475-28	1119	X	X	65-58135-20	1126		X
65-58125-3	1124	X	X	65-65919-4	1108	X	X
65-58126-20	1125	X	X	65-69659-1	1142A	X	X
65-58135-18	1126	X		65-70950-7	1136	X	
65-58135-20	1126		X	65-70950-8	1136		X
65-58135-21	1126	X		65-85379-3	1143	X	
65-58135-30	1126	X	X	65-85379-4	1143		X
65-58143-12	1127	X	X	69-15863-1	1130	X	X
65-65919-4	1108	X	X	69-48930-1	1137	X	X
65-70950-11	1138	X		69-56334-1	1139	X	X
65-70950-12	1138		X	JT8D15BG300		X	X
65-77451-4	1125	X	X				
65-85379-3	1143	X					
65-85379-4	1143		X				
69-28727-8	1133	X	X				
69-48936-1	1137	X	X				
69-56334-1	1139	X	X				
JT8D17BG300 *[1]		X	X				

*[1] ENGINE INTERMIX WITH JT8D-15 PER MC 3510-118

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER SB 80-1038

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (PI)
MODEL 201
ENGINE 1 65-73769-13
ENGINE 2 65-73769-14

AIRLINE (PI)
MODEL 209
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-21950-5	1104	X	
65-21950-3	1104	X	X	65-21950-6	1104		X
65-22249-2	1105	X	X	65-22249-2	1105	X	X
65-22454-9	1106	X	X	65-22454-8	1106	X	X
65-27808-1	1108	P	P	65-45799-1	1109	X	X
65-45798-1	1109	X	X	65-50571-1	1114	X	X
65-46851-1	1111	P	P	65-50571-2 *[1]	1114	X	X
65-46853-1	1113	X	X	65-51534-5	1110	P	P
65-51534-1	1110	X	X	65-52971-1	1117	X	X
65-52261-1	1116	X	X	65-58125-3	1124	X	X
65-52971-1	1117	X	X	65-58135-18	1126	X	
65-55750-1	1121	X		65-58135-20	1126		X
65-55750-2	1121		X	65-58234-3	1128	X	X
65-58125-1	1124	X	X	65-65919-4	1108	X	X
65-58135-12	1126	P	P	65-69659-1	1142A	X	X
65-58135-18	1126	P		65-70950-5	1138	X	
65-58135-20	1126		P	65-70950-6	1138		X
65-58145-4	1127	X	X	65-85379-3	1143	X	
65-58234-10	1128	P	P	65-85379-4	1143		X
65-58234-3	1128	X	X	69-15863-1	1130	X	X
65-65919-4	1108	P	P	69-48936-1	1137	X	X
65-70950-1	1138	P		JT8D9ABG301 *[2]		P	P
65-70950-2	1138		P	JT8D15 *[2]*[5]		P	P
65-70950-5	1138	P		JT8D15A *[5]		P	P
65-70950-6	1138		P				
65-73778-1 (SB 78-1005)		P	P				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D7BG3 *[3]		P	P				
JT8D9BG3 *[3]		P	P				

- *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
 - *[2] REPLACEMENT OF JT8D9 ENGINES WITH JT18D15 PER SB 71-1086
 - *[3] ENGINE INTERMIX PER MC 3510-32K ENGINE SUBSTITUTION PER MC 3510-69K
 - *[4] ENGINE INTERMIX WITH JT8D9A PER MC 3510-125K AND SB 71-1081
 - *[5] ENGINE INTERMIX PER MC 3510-212K AND SB 71-1141
- INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (PI)

MODEL 201, 296

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-3	1101	P	P	65-58135-20	1126		X
65-19592-23*[1]	1101	P		65-58135-46	1126	P	P
65-19592-24*[1]	1101		P	65-58234-3	1128	P	P
65-20996-17*[1]	1103	P	P	65-58234-19	1128	P	P
65-21931-21*[1]	1114A	P	P	65-65919-4	1108	X	X
65-21950-5	1104	X		65-69659-1	1142A	X	X
65-21950-6	1104		X	65-70950-5	1138	P	
65-22249-2	1105	X	X	65-70950-6	1138		P
65-22454-9	1106	X	X	65-70950-7	1138	P	
65-45798-7	1109	P	P	65-70950-8	1138		P
65-46851-4*[1]	1111	P	P	65-85379-3	1143	X	
65-50571-5*[1]	1114	P	P	65-85379-4	1143		X
65-51534-5	1110	P	P	65-15863-1	1130	X	X
65-52971-1	1117	P	P	65-48936-1	1137	X	X
65-54476-3*[1]	1118	P	P	69-50911-2*[1]	1141	P	P
65-55475-33	1119	P	P	JT8D9ABG301*[2]		P	P
65-55749-8*[1]	1120	P	P	JT8D9ABG306*[2]		P	P
65-56646-3*[1]	1122	P	P	JT8D15ABG306*[3]		P	P
65-58125-3	1124	X	X	JT8D15ABG313*[3]		P	P
65-58135-18	1126	X					

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX WITH JT8D-15 PER MC 3510-104, MC 3510-121K, AND SB 71-1080

*[3] ENGINE INTERMIX JT8D-15 PER MC 3510-212K AND SB 71-1141

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (PLU)
 MODEL 2A3
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-51

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45799-1	1109	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-55475-28 *[1]	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12 *[1]	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D9ABG301 *[2]		P	P
JT8D15 *[2]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER MC 3510-121K AND SB 71-1135

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (PRS)
MODEL 25A
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-58135-30	*[1] 1126	X	X
65-19592-24	*[1] 1101		X	65-58135-47	*[1] 1126	X	X
65-20996-24	*[1] 1103	X	X	65-58145-12	*[1] 1127	X	X
65-21931-21	1114A	X	X	65-58234-19	1128	X	X
65-21950-5	1104	X		65-65919-4	1108	X	X
65-21950-6	1104		X	65-69659-1	1142A	X	X
65-22249-2	1105	X	X	65-70950-11	1138	X	
65-25736-2	1107	X	X	65-70950-12	1138		X
65-45798-3	1109	X	X	65-85379-3	1143	X	
65-45798-4	1109	X	X	65-85379-4	1143		X
65-45798-7	1109	X	X	69-15935-1	1131	X	X
65-45798-10	1109	X	X	69-28727-8	1133	X	X
65-45799-148	1109	X	X	69-36670-1	1134	X	X
65-45799-166	1109	X	X	69-37478-1	1135	X	X
65-46851-4	*[1] 1111	X	X	69-45172-1	1136	X	X
65-46852-1	1112	X	X	69-48936-1	1137	X	X
65-46853-2	1113	X	X	69-50911-2	*[1] 1141	X	X
65-46858-131	1138	X		JT8D17ABG300		X	X
65-46858-132	1138		X				
65-50882-1	1115	X	X				
65-51534-5	1110	X	X				
65-52261-2	1116	X	X				
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	X	X				
65-55475-34	*[1] 1119	X	X				
65-55749-8	*[1] 1120	X	X				
65-56646-3	*[1] 1122	X	X				
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-24	1125	X	X				
65-58135-1	1126	X	X				
65-58135-3	1126	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (PT)				AIRLINE (PV)			
MODEL 2V6				MODEL 2E1			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-19684-1	1102	X	X
65-21950-6	1104		X	65-21950-3	1104	X	X
65-22249-3	1105	X	X	65-22249-3	1105	X	X
65-45798-6	1109	X	X	65-22454-8	1106	X	X
65-46851-4	1111	X	X	65-45798-1	1109	X	X
65-51534-9	1110	X	X	65-46853-1	1113	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55475-28	1119	X	X	65-52261-1	1116	X	X
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58135-18	1126	X		65-55750-1	1120	X	
65-58135-20	1126		X	65-55750-2	1120		X
65-58135-30	1126	X	X	65-58125-3	1124	X	X
65-58145-12	1127	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-70950-11	1138	X		65-58145-4	1127	X	X
65-70950-12	1138		X	65-58234-10	1128	X	X
65-85379-3	1143	X		65-58234-3	1128	X	X
65-85379-4	1143		X	65-65919-4	1108	X	X
69-28727-8	1133	X	X	65-69659-10	1142A	X	X
69-48936-1	1137	X	X	65-70950-1	1138	X	
69-56334-1	1139	X	X	65-70950-2	1138		X
JT8D17BG300		X	X	69-15863-1	1130	X	
				69-48936-1	1137		X
				JT8D9BG35 *[1]		X	X

*[1] ENGINE INTERMIX WITH JT8D-15 PER MC 3510-6K AND SB 71-1044 R2

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

BOEING
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (PV)				AIRLINE (PV)			
MODEL 2E1				MODEL 2A3			
ENGINE 1 65-73769-55				ENGINE 1 65-73769-63			
ENGINE 2 65-73769-56				ENGINE 2 65-73769-64			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-3	1105	X	X
65-22454-8	1106	X	X	65-22454-9	1106	X	X
65-45798-1	1109	X	X	65-45798-1	1109	X	X
65-46853-1	1113	X	X	65-46853-1	1113	X	X
65-51534-1	1110	X	X	69-51534-1	1110	X	X
65-52261-1	1116	X	X	65-52261-1	1116	X	X
65-52971-1	1117	X	X	65-52971-1	1117	X	X
65-55750-1	1120	X		65-55750-1	1120	X	
65-55750-2	1120		X	65-55750-2	1120		X
65-58125-1	1124	X	X	65-58125-1	1124	X	X
65-58135-18	1126	X		65-58135-18	1126	X	
65-58135-20	1126		X	65-58135-20	1126		X
65-58145-4	1127	X	X	65-58145-4	1127	X	X
65-58234-10	1128	X	X	65-58234-10	1128	X	X
65-58234-3	1128	X	X	65-58234-3	1128	X	X
65-65919-4	1108	X	X	65-65919-4	1108	X	X
65-69659-1	1142A	X	X	65-69659-1	1142A	X	X
65-70950-1	1138	X		65-70950-1	1138	X	
65-70950-2	1138		X	65-70950-2	1138		X
69-15863-1	1130	X	X	69-15863-1	1130	X	X
69-48936-1 *[1]	1137	X	X	69-48936-1	1137	X	X
JT8D9BG3 *[1]		X	X	JT8D9BG3		X	X

*[1] ENGINE INTERMIX WITH JT8D-15 PER
MC 3510-65K AND SB 71-1044 R2

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(PV)

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

AIRLINE (PW)
 MODEL 275C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

AIRLINE (PW)
 MODEL 214
 ENGINE 1 65-73769-37
 ENGINE 2 65-73769-38

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-3	1104	P	P	65-21950-3	1104	P	P
65-21950-5	1104	P		65-21950-5	1104	P	
65-21950-6	1104		P	65-21950-6	1104		P
65-22249-3	1105	X	X	65-22249-3	1105	X	X
65-22454-8	1106	X	X	65-22454-9	1106	X	X
65-51534-1	1110	X	X	65-51534-1	1110	P	P
65-52971-1	1117	X	X	65-51534-5	1110	P	P
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58135-18	1126	X		65-58125-3	1124	X	X
65-58135-20	1126		X	65-58135-18	1126	X	
65-58234-10	1128	X	X	65-58135-20	1126		X
65-58234-3				65-58234-10	1128	X	X
65-65919-4	1108	X	X	65-58234-3	1128	X	X
65-69659-10	1142A	X	X	65-65919-4	1108	X	X
65-70950-1	1138	P		65-69659-1	1142A	P	P
65-70950-2	1138		P	65-69659-2	1142A	P	P
65-85379-3	1143	X		65-70950-1	1138	P	
65-85379-4	1143		X	65-70950-2	1138		P
69-15863-1	1130	X	X	65-85379-3	1143	X	
69-48936-1	1137	X	X	65-85379-4	1143		X
JT8D9ABG42		P	P	69-15863-1	1130	X	X
JT8D9BG42		P	P	69-48936-1	1137	X	X
				69-56334-1	1139	X	X
				JT8D9ABG301		P	P
				JT8D9ABG300		P	P
				JT8D9ABG42		P	P
				JT8D9BG42		P	P
				JT8D9		P	P
				JT8D9A		P	P
				JT8D15		P	P
				JT8D15A		P	P
				JT8D17		P	P
				JT8D17A		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC7200MK3021

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (PW)
MODEL 275C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	69-15863-1	1130	X	X
65-21950-3	1104	P	P	69-48936-1	1137	X	X
65-21950-5	1104	P		69-56334-1	1139	X	X
65-21950-6	1104		P	JT8D9ABG41		P	P
65-22249-2	1105	P	P	JT8D9ABG42		P	P
65-22249-3	1105	P	P	JT8D9ABG300		P	P
65-22454-8	1106	P	P	JT8D9BG3		P	P
65-22454-9	1106	P	P	JT8D9 *[2]		P	P
65-45798-1	1109	P	P	JT8D9A *[1]*[2]		P	P
65-45798-2	1109	P	P	JT8D15 *[2]		P	P
65-45799-127	1109	P	P	JT8D15A *[2]		P	P
65-46851-3	1111	P	P	JT8D17 *[2]		P	P
65-46853-1	1113	P	P	JT8D17A *[1]*[2]		P	P
65-51534-1	1110	P	P				
65-51534-5	1110	P	P				
65-52261-1	1116	P	P				
65-52971-1	1117	X	X				
65-55750-1	1120	P					
65-55750-2	1120		P				
65-58125-1	1124	P	P				
65-58125-3	1124	P	P				
65-58126-20	1125	P	P				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-21	1126	P					
65-58145-4	1127	P	P				
65-58234-10	1128	X	X				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	P	P				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-77451-3	1125	P	P				
65-85379-3	1140	P					
65-85379-4	1140		P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC 3510-239K AND SB 71-1192
*[2] ENGINE INTERMIX PER MC 7200MK3021

POWER PACK INSTALLATIONS
(PW)

AIRLINE (PW)
MODEL 214
ENGINE 1 65-73769-37
ENGINE 2 65-73769-38

AIRLINE (PW)
MODEL 275
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-2	1105	P	P
65-22454-13	1106	X	X	65-22249-3	1105	P	P
65-45798-1	1109	X	X	65-22454-8	1106	P	P
65-46853-1	1113	X	X	65-22454-9	1106	P	P
65-51534-1	1110	X	X	65-45798-1	1109	X	X
65-52261-1	1116	X	X	65-46853-1	1113	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55750-1	1120	X		65-52261-1	1116	X	X
65-55750-2	1120		X	65-52971-1	1117	X	X
65-57815-1	1140	P	P	65-55475-26	1119	P	P
65-57815-2	1140	P	P	65-55750-1	1121	X	
65-58125-1	1124	X	X	65-55750-2	1121		X
65-58135-12	1126	X	X	65-51815-1	1140	P	P
65-58145-1	1127	X	X	65-51815-2	1140	P	P
65-58145-4	1127	X	X	65-58125-1	1124	P	P
65-58234-3	1128	X	X	65-58125-3	1124	P	P
65-65919-4	1108	X	X	65-58135-12	1126	P	P
65-69659-2	1142A	X	X	65-58135-18	1126	P	
65-73778-1	(SB 78-1005)	X	X	65-58135-20	1126		P
69-28727-1	1133	X	X	65-58145-4	1127	X	X
69-48936-1	1137	X	X	65-58234-10	1128	P	P
JT8D7BG3		X	X	65-58234-3	1128	P	P
				65-65919-4	1108	X	X
				65-69659-2	1142A	P	P

(Continued on next page)

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (PW) (Continued)
MODEL 275
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (QB)
MODEL 296
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-70950-1	1138	P		65-21950-5	1104	X	X
65-70950-2	1138		P	65-21950-6	1104		X
65-70950-5	1138	P		65-22249-2	1105	X	X
65-70950-6	1138		P	65-22454-8	1106	X	X
65-73778-1 (SB 78-1005)		P	P	65-45798-2	1109	X	X
69-15863-1	1130	X	X	65-46851-3	1111	X	X
69-48936-1	1137	X	X	65-51534-5	1110	X	X
JT8D9BG21		P	P	65-58125-3	1124	X	X
JT8D9BG3		P	P	65-58126-20	1125	X	X
JT8D9BG35		P	P	65-58135-18	1126	X	
JT8D9BG9		P	P	65-58135-20	1126		X
JT8D9 *[2]		P	P	65-58135-21	1126	X	
JT8D9A *[2]		P	P	65-65919-4	1108	X	X
JT8D15 *[2]		P	P	65-69659-1	1142A	X	X
JT8D15A *[2]		P	P	65-70950-5	1138	X	
JT8D17 *[2]		P	P	65-70950-6	1138		X
JT8D17A *[2]		P	P	65-77451-4	1125	X	X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D9ABG306*[1]		X	X

*[1] ENGINE INTERMIX WITH JT8D-7/-7A/-7B PER MC 3510-182K AND SB 71-1114

*[2] ENGINE INTERMIX PER MC7200MK3021

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (QC)
MODEL 298C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

AIRLINE (QI)
MODEL 2X9
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[2]	1103	X	X	65-19592-23*[1]	1101	X	
65-21950-3	1104	X	X	65-19592-24*[1]	1101		X
65-22249-3	1105	X	X	65-20996-19*[1]	1103	P	P
65-22454-8	1106	X	X	65-21950-5	1104	X	
65-45798-2	1109	X	X	65-21950-6	1104		X
65-45799-127	1109	X	X	65-22249-2	1105	X	X
65-46851-5	1111	X	X	65-45798-4	1109	P	P
65-51534-1	1110	X	X	65-45798-5	1109	P	P
65-52261-1	1116	X	X	65-45798-10	1109	P	P
65-52971-1	1117	X	X	65-45799-148	1109	P	P
65-58125-3	1124	X	X	65-46851-4*[1]	1111	P	P
65-58126-20	1125	X	X	65-46851-5*[1]	1111	P	P
65-58135-18	1126	X		65-51534-5	1110	X	X
65-58135-20	1126		X	65-52971-1	1117	X	X
65-58135-21	1126	X		65-54476-4*[1]	1118	P	P
65-58234-10	1128	X	X	65-55749-9*[1]	1120	P	P
65-58234-3	1128	X	X	65-58122-2*[1]	1123	P	P
65-65919-4	1108	X	X	65-58125-3	1124	X	X
65-69659-1	1142A	X	X	65-58126-20	1125	P	P
65-70950-3	1138	X		65-58135-18	1126	P	
65-70950-4	1138		X	65-58135-20	1126		X
65-77451-3	1125	X	X	65-58135-21	1126	P	
65-85379-1	1143	X		65-58135-30*[1]	1126	X	X
65-85379-2	1143		X	65-58135-47*[1]	1126	X	X
65-85379-3	1143	X		65-58234-3	1128	X	X
65-85379-4	1143		X	65-65919-4	1108	X	X
69-15863-1	1130	X	X	65-69659-1	1142A	X	X
69-48936-1	1137	X	X	65-70950-11	1138	X	
JT8D15BG42		X	X	65-70950-12	1138		X
				65-77451-4	1125	P	P
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D17BG300		P	P
				JT8D17BG301		P	P

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
PER MC 3540-11K AND SB 80-1020

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(QC)(QI)

737

POWERPLANT BUILDUP

BOEING

COMMERCIAL JET

OVERHAUL MANUAL

AIRLINE (QZ)

MODEL 2M9

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-1*[1]	1103	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
62-22249-1	1105	X	X
65-22454-8	1106	X	X
65-45798-5	1109	X	X
65-45799-147	1109	P	P
65-45799-148	1109	P	P
65-46851-5	1111	X	X
65-51534-1	1110	X	X
65-52971-1	1117	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58126-20	1125	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-21	1126	X	
65-58135-26	1126	X	X
65-58145-12	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-2	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-77451-3	1125	X	X
65-85379-3	1143	X	
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D17BG55		X	X

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1021

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(QZ)

Dec 5/84

71-03-01
Page 1262A

AIRLINE (RG)				AIRLINE (RG)			
MODEL 241				MODEL 2C3			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-1	1105	X	X	65-22249-2	1105	X	X
65-22454-8	1106	X	X	65-45798-5	1109	X	X
65-45798-5	1109	X	X	65-45799-147	1109	P	P
65-45799-147	1109	P	P	65-45799-148			
65-45799-148	1109	P	P	65-46851-5	1111	X	X
65-46851-5	1111	X	X	65-51534-1	1110	X	X
65-51534-1	1110	X	X	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-55475-28	1119	X	X
65-55475-28	1119	X	X	65-58125-3	1124	X	X
65-58125-3	1124	X	X	65-58126-20	1125	X	X
65-58126-20	1125	X	X	65-58135-18	1126	X	
65-58135-18	1126	X		65-58135-20	1126		X
65-58135-20	1126		X	65-58135-21	1126	X	
65-58135-21	1126	X		65-58135-26	1126	X	X
65-58135-26	1126	X	X	65-58145-12	1127	X	X
65-58145-12	1127	X	X	65-58234-10	1128	X	X
65-58234-10	1128	X	X	65-58234-3	1128	X	X
65-58234-3	1128	X	X	65-65919-4	1108	X	X
65-65919-4	1108	X	X	65-69659-2	1142A	X	X
65-69659-11	1142A	X	X	65-70950-11	1138	X	
65-70950-11	1138	X		65-70950-12	1138		X
65-70950-12	1138		X	65-77451-3	1125	X	X
65-77451-3	1125	X	X	65-85379-3	1143	X	
65-85379-3	1143	X		65-85379-4	1143		X
65-85379-4	1143		X	69-28727-8	1133	X	X
69-28727-8	1133	X	X	69-48936-1	1137	X	X
69-48936-1	1137	X	X	JT8D17BG53		X	X
JT8D17BG53		X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (RH)
MODEL 2NO
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-58125-3	1124	X	X
65-19592-24 *[1]	1101		X	65-58126-1	1125	X	X
65-20996-24 *[1]	1103	X	X	65-58126-2	1125	X	X
65-21931-21	1114A	X	X	65-58126-24	1125	X	X
65-21950-5	1104	X		65-58135-1	1126	X	X
65-21950-6	1104		X	65-58135-3	1126	X	X
65-22249-1	1105	X	X	65-58135-18	1126	X	
65-25736-2	1107	X	X	65-58135-20	1126		X
65-45798-4	1109	X	X	65-58135-30 *[1]	1126	X	X
65-45798-7	1109	X	X	65-58135-47 *[1]	1126	X	X
65-45798-10	1109	X	X	65-58145-12 *[1]	1127	X	X
65-45799-148	1109	X	X	65-58234-10	1128	X	X
65-46851-1	1111	X	X	65-58234-19	1128	X	X
65-46852-1	1112	X	X	65-65919-4	1108	X	X
65-46853-2	1113	X	X	65-69659-1	1142A	X	X
65-46858-131	1138	X	X	65-70950-11	1138	X	
65-46858-132	1138		X	65-70950-12	1138		X
65-50571-5	1114	X	X	65-85379-3	1143	X	
65-50882-1	1115	X	X	65-85379-4	1143		X
65-51534-5	1110	X	X	69-15935-1	1131	X	X
65-52261-2	1116	X	X	69-28727-8	1133	X	X
65-52971-1	1117	X	X	69-36670-1	1134	X	X
65-54476-4 *[1]	1118	X	X	69-37478-1	1135	X	X
65-55475-28 *[1]	1119	X	X	69-45172-1	1136	X	X
65-55475-33	1119	X	X	69-48936-1	1137	X	X
65-55749-8	1120	X	X	69-50911-2	1141	X	X
65-56646-3	1122	X	X	69-56334-1	1139	X	X
65-58122-2 *[1]	1123	X	X	69-60143-1	1142	X	X
				JT8D17ABG314			

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

AIRLINE (SA)
 MODEL 244
 ENGINE 1 65-73769-33
 ENGINE 2 65-73769-34

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-14	1106	X	X
65-27808-1	1108	P	P
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1120	X	
65-55750-2	1120		X
65-57815-1	1140	P	P
65-57815-2	1140	P	P
65-58125-1	1124	X	X
65-58135-12	1126	P	P
65-58135-18	1126	P	
65-58135-20	1126		P
65-58145-1	1127	X	X
65-58145-4	1127	X	X
65-58234-10	1128	P	P
65-58234-3	1128	X	X
65-65919-4	1108	P	P
65-69659-1	1142A	X	X
65-70950-1	1138	P	
65-70950-2	1138		P
65-70950-5	1138	P	
65-70950-6	1138		P
65-73778-1	(SB 78-1005)	P	P
69-28727-1	1133	X	X
69-48936-1	1137	X	X
JT8D9BG3		P	P
JT8D9BG9		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

AIRLINE (SA)
 MODEL 244
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	P		65-58234-19	1128	X	X
65-19592-24 *[1]	1101		P	65-65919-4	1108	X	X
65-19592-3	1101	P	P	65-69659-1	1142A	X	X
65-20996-11 *[1]	1103	X	X	65-70950-11	1138	X	
65-21931-21	1114A	X	X	65-70950-12	1138		X
65-21950-5	1104	X		65-85379-3	1143	X	
65-21950-6	1104		X	65-85379-4	1143		X
65-22249-3	1105	X	X	69-28727-8	1133	X	X
65-22454-25	1106	X	X	69-48936-1	1137	X	X
65-45798-6	1109	X	X	69-50911-2 *[1]	1141	X	X
65-45798-7	1109	X	X	69-56334-1	1139	X	X
65-45798-9	1109	X	X				
65-45799-148	1109	X	X	JT8D17ABG300 *[2]		P	P
65-45799-165	1109	X	X	JT8D17BG300 *[2]		P	P
65-45799-166	1109	X	X	JT8D17A *[3]		P	P
65-45799-170	1109	P	P	JT8D17 *[3]		P	P
65-46851-4 *[1]	1111	X	X	JT8D15A *[3]		P	P
65-50882-3 *[1]	1115	X	X	JT8D15 *[3]		P	P
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-54476-2 *[1]	1118	P	P				
65-54476-4 *[1]	1118	P	P				
65-55475-33 *[1]	1119	X	X				
65-55475-34 *[1]	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-3 *[1]	1122	X	X				
65-58122-2 *[1]	1123	P	P				
65-58125-3	1124	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-30 *[1]	1126	P	P				
65-58135-31	1126	X	X				
65-58135-37	1126	P	P				
65-58135-48 *[1]	1126	P	P				
65-58145-12 *[1]	1127	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

- *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
- *[2] ENGINE INTERMIX PER MC 3510-161K AND SB 71-1085
- *[3] ENGINE INTERMIX PER MC 3510-222K AND SB 71-1173

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (SE)
MODEL 229
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-1	1105	X	X
65-22454-9		X	X
65-45799-1	1109	X	X
65-46851-4	1111	X	X
65-51534-1	1110	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-9	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D15BG42		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(SE)

737

POWERPLANT BUILDUP

BOEING

COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (SD)				AIRLINE (SH)			
MODEL 2J8C				MODEL 2K6			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-2	1105	X	X
65-22249-3	11-5	X	X	65-22454-8	1106	X	X
65-22454-8	1106	X	X	65-46851-4	1111	X	X
65-45798-5	1109	X	X	65-51534-1	1110	P	P
65-45799-147	1109	X	X	65-51534-5	1110	P	P
65-45799-148	1109	X	X	65-52971-1	1117	X	X
65-46851-5	1111	X	X	65-55475-28	1119	X	X
65-51534-1	1110	X	X	65-58125-3	1124	X	X
65-52971-1	1117	X	X	65-58135-18	1126	X	
65-58125-3	1124	X	X	65-58135-20	1126		X
65-58126-20	1125	X	X	65-58145-12	1127	X	X
65-58135-18	1126	X		65-58234-10	1128	X	X
65-58135-20	1126		X	65-58234-3	1128	X	X
65-58135-21	1126	X		65-65919-4	1108	X	X
65-58135-26	1126	X	X	65-69659-1	1142A	X	X
65-58234-10	1128	X	X	65-70950-5	1138	P	
65-58234-3	1128	X	X	65-70950-6	1138		P
65-65919-4	1108	X	X	65-70950-7	1138	P	
65-69659-1	1142A	X	X	65-70950-8	1138		P
65-70950-11	1138	X		65-85379-3	1143	X	
65-70950-12	1138		X	65-85379-4	1143		X
65-77451-3	1125	X	X	69-28727-1	1133	X	X
65-85379-3	1143	X		69-48936-1	1137	X	X
65-85379-4	1143		X	JT8D15BG42		X	X
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D17BG55		X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (SD)(SH)

Jan 5/81

71-03-01
 Page 1265

OVERHAUL MANUAL

AIRLINE (SN)
MODEL 229C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

AIRLINE (SN)
MODEL 229
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-3	1104	P	P	65-21950-3	1104	P	P
65-21950-5	1104	P		65-21950-5	1104	P	
65-21950-6	1104		P	65-21950-6	1104		P
65-22249-1	1105	X	X	65-22249-1	1106	X	X
65-22454-9	1106	X	X	65-22454-9	1106	X	X
65-46851-4	1111	X	X	65-46851-4	1111	X	X
65-51534-1	1110	P	P	65-51534-1	1110	X	X
65-51534-5	1110	P	P	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-58125-3	1124	X	X
65-58125-3	1124	X	X	65-58135-18	1126	X	
65-58135-18	1126	X		65-58135-20	1126		X
65-58135-20	1126		X	65-58234-10	1128	X	X
65-58234-10	1128	X	X	65-58234-9	1128	X	X
65-58234-12	1128	X	X	65-65919-4	1108	X	X
65-58234-9	1128	X	X	65-69659-1	1142A	X	X
65-65919-4	1108	X	X	65-70950-7	1138	X	
65-69659-1	1142A	X	X	65-70950-8	1138		X
65-70950-7	1138	X		65-85379-3	1143	X	
65-70950-8	1138		X	65-85379-4	1143		X
65-85379-3	1143	X		69-15863-1	1130	X	X
65-85379-4	1143		X	69-48936-1	1137	X	X
69-15863-1	1130	X	X	JT8D15BG42		P	P
69-48936-1	1137	X	X	JT8D15*[1]		P	P
JT8D15BG42		P	P	JT8D-15A*[1]		P	P
JT8D-15*[1]		P	P				
JT8D-15A*[1]		P	P				

*[1] ENGINE INTERMIX PER
MC 3510-197K

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (SQ)

MODEL 112

ENGINE 1 65-73769-45

ENGINE 2 65-73769-46

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-8	1106	X	X
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1120	X	
65-55750-2	1120		X
65-57815-1	1140	X	X
65-57815-2	1140	X	X
65-58125-1	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-4	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-1	1138	X	
65-70950-2	1138		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D9BG3		P	P
JT8D9BG9		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

AIRLINE (SV)				AIRLINE (SV) (CONT)			
MODEL 275							
ENGINE 1 65-73769-51							
ENGINE 2 65-73769-52							
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-70950-3	1138	X	
65-20996-1*[1]	1103	X	X	65-70950-4	1138		X
65-21950-3	1104	P	P	65-70950-7	1138	P	
65-21950-5	1104	P		65-70950-8	1138		P
65-21950-6	1104		P	65-77451-2	1125	P	P
65-22249-3	1105	X	X	65-77451-3	1125	P	P
65-22454-24	1106	P	P	65-77451-4	1125	P	P
65-22454-8	1106	P	P	65-85379-1	1143	P	
65-45798-1	1109	P	P	65-85379-2	1143		P
65-45798-2	1109	X	X	65-85379-3	1143	P	
65-45799-127	1109	P	P	65-85879-4	1140		P
65-46851-5	1111	X	X	65C13071-2	1130	P	P
65-46853-1	1113	X	X	69-15863-	1130	X	X
65-51534-1	1110	X	X	69-48936-1	1137	X	X
65-51534-5	1110	X	X	69-56334-1	1139	X	X
65-52261-1	1116	X	X	JT8D15BG28		P	P
65-52971-1	1117	X	X	JT8D15BG48		P	P
65-55750-1	1120	P		JT8D15BG304		P	P
65-55750-2	1120		P				
65-58125-3	1124	X	X				
65-58126-20	1125	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-21	1126	X					
65-58145-14	1127	P	P				
65-58145-4	1127	P	P				
65-58234-10	1128	P	P				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				

*[1] INCLUDES ENGINE STARTER VALVE OPERATING INDICATION LIGHT PRESSURE SWITCH PER SB 80-1006

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (SV)
 MODEL 268C
 ENGINE 1 65-73769-75
 ENGINE 2 65-73769-76

AIRLINE (TA)
 MODEL 2A1
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-20996-10	1103	X	X
65-20996-1*[1]	1103	X	X	65-21950-5	1104	X	
65-21950-3	1104	X	X	65-21950-6	1104		X
65-22249-3	1105	X	X	65-22249-3	1105	X	X
65-22454-8	1106	X	X	65-45798-1	1109	X	X
65-45798-1	1109	X	X	65-46851-4	1111	X	X
65-45798-2	1109	X	X	65-51534-1	1110	P	P
65-45799-127	1109	X	X	65-51534-5	1110	P	P
65-46851-5	1111	X	X	65-52971-1	1117	X	X
65-46853-1	1113	X	X	65-55475-28	1119	X	X
65-51534-1	1110	X	X	65-58125-3	1124	X	X
65-52261-1	1116	X	X	65-58135-18	1126	X	
65-52971-1	1117	X	X	65-58135-20	1126		X
65-55750-1	1120	X		65-58135-26	1126	X	X
65-55750-2	1120		X	65-58145-12	1127	X	X
65-58125-3	1124	X	X	65-58234-3	1128	X	X
65-58126-20	1125	X	X	65-65919-4	1108	X	X
65-58135-18	1126	X		65-69659-1		X	X
65-58135-20	1126		X	65-70950-11	1138	X	
65-58135-21	1126	X		65-70950-12	1138		X
65-58145-4	1127	X	X	65-85379-3	1143	X	
65-58234-10	1128	X	X	65-85379-4	1143		X
65-58234-5	1128	X	X	69-28727-8	1133	X	X
65-65919-4	1108	X	X	69-48936-1	1137	X	X
65-69659-1	1142A	X	X				
65-70950-3	1138	X		JT8D17BG42		P	P
65-70950-4	1138		X	JT8D17BG301		P	P
65-77451-2	1125	X	X				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
69-56334-1	1139	X	X				
JT8D15BG28		X	X				

*[1] INCLUDES ENGINE STARTER VALVE OPERATING INDICATION LIGHT PRESSURE SWITCH PER SB 80-1006

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (SV) (IE)

AIRLINE (TA)
MODEL 222
ENGINE 1 65-73769-5
ENGINE 2 65-73769-6

AIRLINE (TA)
MODEL 244
ENGINE 1 65-73769-33
ENGINE 2 65-73769-34

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-1	1105	X	X	65-22249-2	1105	X	X
65-22454-8	1106	X	X	65-22454-14	1106	X	X
65-25736-3	1107	X	X	65-45798-1	1109	X	X
65-45798-1	1109	X	X	65-46853-1	1113	X	X
65-46853-1	1113	X	X	65-51534-1	1110	X	X
65-50571-2	1114	X	X	65-52261-1	1116	X	X
65-51534-1	1110	X	X	65-52971-1	1117	X	X
65-52261-1	1116	X	X	65-55750-1	1120	X	
65-52971-1	1117	X	X	65-55750-2	1120		X
65-55750-1	1121	X		65-58125-1	1124	X	X
65-55750-2	1121		X	65-58135-12	1126	X	X
65-58125-1	1124	X	X	65-58145-1	1127	X	X
65-58135-18	1126	X		65-58145-4	1127	X	X
65-58145-4	1127	X	X	65-58234-3	1128	X	X
65-58234-11	1128	X	X	65-65919-4	1108	X	X
65-58234-9	1128	X	X	65-69659-1	1142A	X	X
65-69659-5	1142A	P	P	65-73778-1		X	X
65-70950-1	1138	X		69-28727-1	1133	X	X
65-70950-2	1138		X	69-48936-1	1137	X	X
65-70950-5	1138	X					
65-70950-6	1138		X	JT8D9BG3 *[1]		P	P
69-15863-1	1130	X	X	JT8D9 *[1]		P	P
69-48936-1	1137	X	X	JT8D9A *[1]		P	P
				JT8D17 *[1]		P	P
JT8D7BG3 *[1]		P	P				
JT8D9 *[1]		P	P				
JT8D9A *[1]		P	P				
JT8D17 *[1]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC 7200MK3013

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (TAF)
 MODEL 2Z6
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (TH)
 MODEL 2P5
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-19592-23 *[1]	1101	X	
65-19592-24 *[1]	1101		X	65-19592-24 *[1]	1101		X
65-20996-1 *[2]	1103	X	X	65-21950-5	1104	X	
65-22249-2	1105	X	X	65-21950-6	1104		X
65-45798-10 *[1]	1109	X	X	65-22249-2	1105	X	X
65-45799-170	1109	X	X	65-45798-10	1109	X	X
65-46851-4 *[1]	1111	X	X	65-45799-1	1109	X	X
65-51534-5	1110	X	X	65-46851-4 *[1]	1111	X	X
65-52971-1	1117	X	X	65-51534-5	1110	X	X
65-54476-4 *[1]	1118	X	X	65-52971-1	1117	X	X
65-55749-9 *[1]	1120	X	X	65-54476-4 *[1]	1118	X	X
65-58122-2 *[1]	1123	X	X	65-55749-9 *[1]	1120	X	X
65-58125-3	1124	X	X	65-58122-2 *[1]	1123	X	X
65-58135-18	1126	X		65-58125-3	1124	X	X
65-58135-20	1126		X	65-58135-18	1126	X	
65-58135-47 *[1]	1126	X	X	65-58135-20	1126		X
65-58234-3	1128	X	X	65-58135-47 *[1]	1126	X	X
65-65919-4	1108	X	X	65-58234-3	1128	X	X
65-69659-1	1142A	X	X	65-65919-4	1108	X	X
65-69659-13	1142A	X	X	65-69659-1	1142A	X	X
65-70950-7	1139	X		65-69659-13	1142A	X	X
65-70950-8	1138		X	65-70950-7	1138	X	
65-85379-3	1143	X		65-70950-8	1138		X
65-85379-4	1143		X	65-85379-3	1143	X	
69-15863-1	1130	X	X	65-85379-4	1143		X
69-48936-1	1137	X	X	69-15863-1	1130	X	X
JT8D15ABG300		X	X	69-48936-1	1137	X	X
				JT8D15BG300		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1016

POWER PACK INSTALLATIONS
 (TAF) (TH)

AIRLINE (TH)
 MODEL 2P5
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (TJ)
 MODEL 2R8C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-23

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1 *[1]	1103	X	X	65-21950-5	1104	X	
65-21950-5	1104	X		65-21950-6	1104		X
65-21950-6	1104		X	65-22249-3	1105	X	X
65-22249-2	1105	X	X	65-22454-8	1106	X	X
65-46851-4	1111	X	X	65-46851-4	1111	X	X
65-51534-1	1110	P	P	65-51534-5	1110	X	X
65-51534-5	1110	P	P	65-52971-1	1117	X	X
65-52971-1	1117	X	X	65-58125-3	1124	X	X
65-58125-3	1124	X	X	65-58135-18	1126	X	
65-58135-18	1126	X		65-58135-20	1126		X
65-58135-20	1126		X	65-58135-30	1126	X	X
65-58234-3	1128	X	X	65-65919-4	1108	X	X
65-65919-4	1108	X	X	65-69659-1	1142A	X	X
65-69659-1	1142A	X	X	65-70950-11	1138	X	
65-70950-7	1138	X		65-70950-12	1138		X
65-70950-8	1138		X	65-85379-3	1142	X	
65-85379-3	1140	X		65-85379-4	1143		X
65-85379-4	1143		X	69-15863-1	1130	X	X
69-15863-1	1130	X	X	69-48936-1	1137	X	X
69-48936-1	1137	X	X				
JT8D15BG301		X	X	JT8D17BG306		X	X
JT8D15BG42 (OPT)		X	X				
JT8D9ABG42 (OPT)		X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1027

OVERHAUL MANUAL

AIRLINE (TM)				AIRLINE (TM)			
MODEL 2B1C				MODEL 2B1			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-51			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-52			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-2	1104	X	X	65-21950-3	1104	X	X
65-21950-3	1104	X	X	65-22249-2	1105	X	X
65-22249-2	1105	X	X	65-22454-8	1106	X	X
65-22454-8	1106	X	X	65-45798-1	1109	X	X
65-46853-1	1113	X	X	65-46853-1	1111	X	X
65-51534-1	1110	X	X	65-51534-1	1110	X	X
65-52261-1	1116	X	X	65-52261-1	1116	X	X
65-55750-1	1120	X		65-52971-1	1117	X	X
65-55750-2	1120		X	65-55750-1	1120	X	
65-58125-3	1124	X	X	65-55750-2	1120		X
65-58135-18	1126	X		65-58125-1	1124	P	P
65-58135-20	1126		X	65-58125-3	1124	P	P
65-58145-4	1127	X	X	65-58135-18	1126	X	
65-58234-10	1128	X	X	65-58135-20	1126		X
65-58234-3	1128	X	X	65-58145-4	1127	X	X
65-65919-4	1108	X	X	65-58234-10	1128	X	X
65-69659-1	1142A	X	X	65-58234-3	1128	X	X
65-70950-1	1138	X		65-65919-4	1108	X	X
65-70950-2	1138		X	65-69659-1	1142A	X	X
65-70950-5	1138	X		65-70950-1	1138	X	
65-70950-6	1138		X	65-70950-2	1138		X
69-15863-1	1130	X	X	65-70950-5	1138	X	
69-48936-1	1137	X	X	65-70950-6	1138		X
JT8D9BG21		X	X	69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D9BG21		P	P
				JT8D9BG9		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (TP)

MODEL 282

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-70950-11	1138	X	
65-19592-24 *[1]	1101		X	65-70950-12	1138		X
65-20996-19 *[1]	1103	X	X	65-85379-3	1143	X	
65-21931-22	1114A	X	X	65-85379-4	1143		X
65-21950-5	1104	X		69-28727-8	1133	X	X
65-21950-6	1104		X	69-36670-2 *[1]	1134	X	X
65-22249-1	1105	X	X	69-48936-1	1137	X	X
65-45798-4	1109	X	X	69-50911-2 *[1]	1141	X	X
65-45798-6	1109	X	X	JT8D17ABG303		P	P
65-45798-7	1109	X	X	JT8D17ABG305		P	P
65-45798-10 *[1]	1109	X	X				
65-45799-148 *[1]	1109	X	X				
65-45799-165	1109	X	X				
65-45799-166	1109	X	X				
65-46851-4 *[1]	1111	X	X				
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X				
65-55475-33 *[1]	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-2 *[1]	1122	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58135-47 *[1]	1126	X	X				
65-58135-30 *[1]	1126	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58145-12 *[1]	1127	X	X				
65-58234-17	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (TP)
MODEL 282C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-85379-3	1143	X	
65-19592-24 *[1]	1101		X	65-85379-4	1143		X
65-20996-19 *[1]	1103	X	X	69-28727-8	1133	X	X
65-21931-22	1114A	X	X	69-36670-2 *[1]	1134	X	X
65-21950-5	1104	X		69-50911-2 *[1]	1141	X	X
65-21950-6	1104		X	JT8D17ABG303		X	X
65-22249-1	1105	X	X				
65-45798-4	1109	X	X				
65-45798-6	1109	X	X				
65-45798-7	1109	X	X				
65-45798-10*[1]	1109	X	X				
65-45799-148	1109	X	X				
65-45799-165	1109	X	X				
65-45799-166	1109	X	X				
65-45799-170	1109	X	X				
65-46851-4*[1]	1111	X	X				
65-51534-8	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X				
65-55475-33 *[1]	1119	X	X				
65-55475-34	1119	X	X				
65-55749-8 *[1]	1120	X	X				
65-56646-2 *[1]	1122	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58135-30 *[1]	1126	X	X				
65-58135-47 *[1]	1126	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58145-12 *[1]	1127	X	X				
65-58234-17	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-70950-11	1138	X					
65-70950-12	1138		X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

POWER PACK INSTALLATIONS
(TP)

AIRLINE (TP)

MODEL 2K9

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-22249-2	1105	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-57815	1140	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-5	1138	X	
65-70950-6	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17BG301		P	P
JT8D17 *[1]		P	P
JT8D17A *[1]		P	P

*[1] ENGINE INTERMIX PER MC 7200MK3041

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737



POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (TQ)

MODEL 2M8, 2Q9

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-20996-1 *[2]		P	P
65-21950-5	1104	X	
65-21950-6	1104		X
64-22249-1	1105	X	X
65-22454-8	1106	X	X
65-51534-1	1110	P	P
65-51534-5	1110	P	P
65-52971-1	1117	X	X
65-55475-28	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	P	
65-70950-8	1138		P
65-85379-3	1143		X
65-85379-4	1143		X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D9ABG42 *[1]		P	P
JT8D15BG42 *[1]		P	P

*[1] ENGINE INTERMIX PER MC 3510-65K

*[2] INCLUDES STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER SB 80-1037

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (TS)

MODEL 297
 ENGINE 1 65-73769-61
 ENGINE 2 65-73769-62

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	P	P
65-21950-3	1104	X	X
65-22249-2	1105	X	X
65-22454-9	1106	X	X
65-45798-1	1109	X	X
65-46853-1	1113	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1121	X	
65-55750-2	1121		X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58234-10	1128		
65-65919-4	1108	X	X
65-69659-10	1142A	X	X
65-70950-1	1138	X	
65-70950-2	1138		X
65-85379-3	1143	P	
65-85379-4	1143		P
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D9BG3		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (TS)

MODEL 284, 297
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1	1103	P	P	JT8D-9A *[4]		P	P
65-20996-11	1103	P	P	JT8D-7 *[4]		P	P
65-21950-5	1104	X		JT8D-7A *[4]		P	P
65-21950-6	1104		X	JT8D-7B *[4]		P	P
65-22249-2	1105	X	X				
65-22454-8	1106	X	X	*[1] ENGINE INTERMIX WITH JT8D-15 PER MC 3510-121K AND SB 71-1095			
65-51534-1	1110	X	X				
65-52971-1	1117	X	X				
65-58125-3	1124	X	X	*[2] ENGINE RETROFIT AND INTERMIX WITH JT8D-15 PER MC 3510-184K			
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58234-9	1128	X	X	*[3] ENGINE RETROFIT AND INTERMIX PER MC 3530-15K AND SB 71-1137			
65-65919-4	1108	X	X				
65-69695-1	1142A	X	X				
65-70950-5	1138	X		*[4] ENGINE INTERMIX JT8D-7, -7A, -7B WITH JT8D-9, -9A PER MC 3510-172K AND SB 71-1189			
65-70950-6	1138		X				
65-85379-3	1143	X					
65-85379-4	1143		X				
65-15863-1	1130	X	X				
65-48936-1	1137	X	X				
JT8D9ABG301 *[1]		P	P				
*[2]							
JT8D9ABG42 *[1]		P	P				
JT8D-9A *[3]		P	P				
JT8D-15 *[3]		P	P				
JT8D-9 *[3]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



AIRLINE (TS)

MODEL 2S5C
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45798-5	1109	X	X
65-46851-5	1111	X	X
65-51534-5	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55475-28	1119	X	X
65-58126-20	1125	X	X
65-58135-30	1126	X	X
65-58135-21	1126	X	
65-58145-15	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-77451-4	1125	X	X
65C13071-3	1133	X	X
69-28727-8	1133	X	X
69-56334-1	1139	X	X
JT8D17BG300		P	P
JT8D17 *[1]		P	P
JT8D9/9A *[1]		P	P

*[1] ENGINE INTERMIX PER MC 7200MK3039 AND SB 71-1210

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (TU)				AIRLINE (TU)			
MODEL 2H3				MODEL 2H3C			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[1]	1103	X	X	65-20996-1*[1]	1103	X	X
65-21950-5	1104	X		65-21950-5	1104	X	
65-21950-6	1104		X	65-21950-6	1104		X
65-22249-2	1105	X	X	65-22249-2	1105	P	P
65-51534-5	1110	X	X	65-22249-3	1105	P	P
65-52971-1	1117	X	X	65-22454-8	1105	X	X
65-58125-3	1124	X	X	65-51534-5	1110	X	X
65-58135-18	1126	X		65-52971-1	1117	X	X
65-58135-20	1126		X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-1	1142A	X	X	65-58234-3	1128	X	X
65-70950-11	1138	P		65-65919-4	1108	X	X
65-70950-12	1138		P	65-69659-1	1142A	X	X
65-85379-3	1143	X		65-70950-5	1138	X	
65-85379-4	1143		X	65-70950-6	1138		X
69-15863-1	1130	X	X	65-85379-3	1143	X	
69-48936-1	1137	X	X	65-85379-4	1143		X
JT8D9ABG42		X	X	69-15863-1	1130	X	X
				69-48936-1	1137	X	X
				JT8D9ABG301		P	P
				JT8D9ABG42		P	P

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH
 PER MC 3540-11K AND SB 80-1023

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP

BOEING

COMMERCIAL JET
OVERHAUL MANUAL

AIRLINE (TX)				AIRLINE (TZ)			
MODEL 2A3				MODEL 2A9C			
ENGINE 1 65-73769-63				ENGINE 1 65-73769023			
ENGINE 2 65-73769-64				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE 1	ENGINE 2	INSTALLATION	FIGURE	ENGINE 1	ENGINE 2
65-19684-1	1102	X	X	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-2	1104	X	X
65-22249-3	1105	X	X	65-21950-3	1104	X	X
65-22454-9	1106	X	X	65-22249-2	1105	X	X
65-45798-1	1109	X	X	65-22454-8	1106	X	X
65-46853-1	1113	X	X	65-45798-1	1108	X	X
65-51534-1	1110	X	X	65-46853-1	1113	X	X
65-52261-1	1116	X	X	65-51534-1	1110	X	X
65-52971-1	1117	X	X	65-52261-1	1116	X	X
65-55750-1	1120	X		65-52971-1	1117	X	X
65-55750-2	1120		X	65-55750-1	1120	X	
65-58125-1	1124	X	X	65-55750-2	1120		X
65-58135-18	1126	X		65-58125-1	1124	X	X
65-58135-20	1126		X	65-58135-18	1126	X	
65-58145-4	1127	X	X	65-58135-20	1126		X
65-58234-10	1128	X	X	65-58145-4	1127	X	X
65-58234-3	1128	X	X	65-58234-10	1128	X	X
65-65919-4	1108	X	X	65-58234-3	1128	X	X
65-69659-1	1142A	X	X	65-65919-4	1108	X	X
65-70950-1	1138	X		65-69659-2	1142A	X	X
65-70950-2	1138		X	65-70950-1	1138	X	
69-15863-1	1130	X	X	65-70950-2	1138		X
69-48936-1	1137	X	X	69-15863-1	1130	X	
JT8D9BG3		X	X	69-48936-1	1137		X
				JT8D9BG9		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (TX)(TZ)

Jan 5/81

71-03-01
 Page 1275

BOEING
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (TZ)
 MODEL 2A9
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (UA)
 MODEL 222
 ENGINE 1 65-73769-5
 ENGINE 2 65-73769-6

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-3	1104	X	X	65-19684-1	1102	X	X
65-22249-3	1105	X	X	65-21950-3	1104	X	X
65-22454-8	1106	X	X	65-22249-1	1105	X	X
65-45798-2	1109	X	X	65-22454-8	1106	X	X
65-45799-127	1109	X	X	65-25736-3	1107	X	X
65-46851-3	1111	X	X	65-27808-1	1108	P	P
65-51534-1	1110	X	X	65-45798-1	1109	X	X
65-52971-1	1117	X	X	65-46853-1	1113	X	X
65-58125-3	1124	X	X	65-50571-2	1114	P	P
65-58126-20	1125	X	X	65-51534-1	1110	X	X
65-58135-18	1126	X		65-52261-1	1116	X	X
65-58135-20	1126		X	65-52971-1	1117	X	X
65-58135-21	1126	X		65-55750-1	1121	X	
65-58234-10	1128	X	X	65-55750-2	1121		X
65-58234-3	1128	X	X	65-57815-1	1140	P	P
65-65919-4	1108	X	X	65-57815-2	1140	P	P
65-69659-2	1142A	X	X	65-58125-1	1124	X	X
65-70950-5	1138	X		65-58125-2	1124	P	P
65-70950-6	1138		X	65-58135-12	1126	P	P
65-77451-3	1125	X	X	65-58135-18	1126	P	
65-85379-3	1143	X		65-58135-20	1126		P
65-85379-4	1143		X	65-58145-4	1127	X	X
69-15863-1	1130	X	X	65-58234-11	1128	P	P
69-48936-1	1137	X	X	65-58234-9	1128	X	X
JT8D9ABG42		X	X	65-65919-4	1108	P	P
				65-69659-5	1142A	P	P
				65-70950-1	1138	P	
				65-70950-2	1138		P
				65-70950-5	1138	P	
				65-70950-6	1138		P
				65-73778-1	(SB 78-1005)	P	P
				69-15863-1	1130	X	X
				69-48936-1	1137	P	P
				JT8D7BG22		P	P
				JT8D7BG3		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (UA)
MODEL 291, 2A1
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-3		P	P	JT8D17BG301		P	P
65-19592-23		P		JT8D9ABG42		P	P
65-19592-24			P	JT8D9ABG301 *[1]		P	P
65-20996-10	1103	P	P	JT9D17 *[1]*[3]*[5]		P	P
65-21950-5	1104	P	P	JT8D9 *[2]*[3]		P	P
65-22249-2	1105	P	P	JT8D9A *[1]*[2]*[3]		P	P
65-22249-3	1105	P	P	*[4]			
65-22454-9	1106	P	P	JT8D7 *[2]]		P	P
65-46851-4	1111	P	P	JT8D7A *[2]		P	P
65-55475-28	1119	P	P	JT8D7B *[2]		P	P
65-58125-3	1124	P	P	JT8D15 *[4]*[5]		P	P
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-26	1126	P	P				
65-58135-30	1126	P	P				
65-58145-12	1127	P	P				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-70950-11	1138	P					
65-70950-12	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-85379-3	1143	P					
65-85379-4	1143		P				
69-15863-1	1130	X	X				
69-28727-8	1133	X	X				
69-48936-1	1137	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

P = PARTIAL USAGE

- *[1] ENGINE INTERMIX PER MC 3510-213K
- *[2] ENGINE INTERMIX PER MC 3510-223K
- *[3] ENGINE INTERMIX PER MC 3510-224K
- *[4] ENGINE INTERMIX PER MC 3510-227K
- *[5] ENGINE INTERMIX PER MC 3510-228K

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (UC)
 MODEL 2K6
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

AIRLINE (UC)
 MODEL 2E3
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-21950-5	1104	X	
65-21950-6	1104		X	65-21950-6	1104		X
65-22249-1	1105	X	X	65-22249-1	1105	X	X
65-22454-8	1106	X	X	65-46851-4	1111	X	X
65-45799-1	1109	X	X	65-51534-5	1110	X	X
65-46851-4	1111	X	X	65-52971-1	1117	X	X
65-51534-5	1110	X	X	65-55475-28	1119	X	X
65-55475-28	1119	X	X	65-58125-3	1124	X	X
65-58125-3	1124	X	X	65-58135-18	1126	X	
65-58135-18	1126	X		65-58135-20	1126		X
65-58135-20	1126		X	65-58145-12	1127	X	X
65-58145-12	1127	X	X	65-65919-4	1108	X	X
65-65919-4	1108	X	X	65-70950-7	1138	X	
65-69659-1	1142A	X	X	65-70950-8	1138		X
65-70950-7	1138	X		65-85379-3	1143	X	
65-70950-8	1138		X	65-85379-4	1143		X
65-85379-3	1143	X		69-28727-8	1133	X	X
65-85379-4	1143		X	69-48936-1	1137	X	X
69-28727-8	1133	X	X	69-56334-1	1139	X	X
69-48936-1	1137	X	X	JT8D15BG305		X	X
JT8D15BG301		P	P				
JT8D17 *[1]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] ENGINE INTERMIX WITH JT8D15 PER MC 3510-124K AND SB 71-1104

AIRLINE (UP)
MODEL 297
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (UH)
MODEL 284, 297
ENGINE 1 65-73769-23
ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-1*[2]	1103	X	X	65-20996-1*[1]	1103	X	X
65-21950-5	1104	X	X	65-21950-5	1104	X	
65-45799-1	1109	X	X	65-21950-6	1104		X
65-51534-5	1110	X	X	65-22249-1	1105	X	X
65-58125-3	1124	X	X	65-22454-8	1106	X	X
65-58135-18	1126	X		65-45798-5	1109	X	X
65-58135-20	1126		X	65-46851-5	1111	X	X
65-65919-4	1108	X	X	65-51534-5	1110	X	X
65-69659-1	1142A	X	X	65-52971-1	1117	X	X
65-70950-5	1138	X		65-55475-28	1119	X	X
65-70950-6	1138		X	65-57815	1140	X	X
65-85379-3	1143	X		65-58125-3	1124	X	X
65-85379-4	1143		X	65-58126-20	1125	X	X
69-15863-1	1130	X	X	65-58135-18	1126	X	
69-48936-1	1137	X	X	65-58135-20	1126	X	
				65-58135-21	1126		X
				65-58135-30	1126	X	
				65-58145-15	1127	X	X
				65-58234-3	1128	X	X
				65-65919-4	1108	X	X
				65-69659-1	1142A	X	X
				65-70950-11	1138	X	
				65-70950-12	1138		X
				65-77451-4	1125		X
				65-85379-3	1143	X	X
				65-85379-4	1143	X	
				65C13071-3	1130		X
				69-28727-8	1133	X	X
				69-48936-1	1137	X	X
				69-56334-1	1139	X	X
				JT8D17BG304		X	X

*[1] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1007

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1015

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

POWER PACK INSTALLATIONS
(UP)(UH)

POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (US)				AIRLINE (VM)			
MODEL T-43A				MODEL 2H7C			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-23			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-24			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-19684-1	1102	X	X
65-21950-3	1104	X	X	65-21950-3	1104	X	X
65-22249-2	1105	X	X	65-22249-1	1105	X	X
65-22454-8	1106	X	X	65-22454-8	1106	X	X
65-45798-1	1109	X	X	65-45798-2	1109	X	X
65-46853-1	1113	X	X	65-45799-127	1109	P	P
65-51534-1	1110	X	X	65-45799-134	1109	P	P
65-52261-1	1116	X	X	65-46851-5	1111	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55750-1	1120	P		65-52971-1	1117	X	X
65-55750-2	1120		P	65-55475-26	1119	X	X
65-58125-3	1124	X		65-55750-1	1120	X	
65-58135-18	1126	X		65-55750-2	1120		X
65-58135-20	1126		X	65-58125-3	1124	X	X
65-58145-4	1127	P	P	65-58126-20	1125	X	X
65-58234-10	1128	X	X	65-58135-18	1126	X	
65-58234-3	1128	X	X	65-58135-20	1126		X
65-65919-4	1108	X	X	65-58135-21	1126	X	
65-69659-1	1142A	X	X	65-58234-10	1128	X	X
65-70950-1	1138	X		65-58234-3	1128	X	X
65-70950-2	1138		X	65-65919-4	1108	X	X
65-85379-1	1143	P		65-69659-1	1142A	X	X
65-85379-2	1143		P	65-70950-3	1138	X	
65-85379-3	1143	P		65-70950-4	1138		X
65-85379-4	1143		P	65-77451-2	1125	X	X
69-15863-1	1130	X	X	65-85379-1	1143	X	
69-48936-1	1137	X	X	65-85379-2	1143		X
JT8D9ABG21		P	P	69-15863-1	1130	X	X
JT8D9BG21		P	P	69-48936-1	1137	X	X
				JT8D15BG16		X	X
				JT8D-9 *[1]		X	X
				JT8D-9A *[1]		X	X
				JT8D-15A *[1]		X	X
				JT8D-17 *[1]		X	X
				JT8D-17A *[1]		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] INTERMIX ENGINES PER SB 71-1351.

POWER PACK INSTALLATIONS
(US) (VM)

POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (VM)
 MODEL 2K9
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		JT8D15A *[2]		P	P
65-19592-24 *[1]	1101		X	JT8D15 *[2]		P	P
65-20996-24	1103	X	X	JT8D17 *[2]		P	P
65-21950-5	1104	X		JT8D9 *[2]		P	P
65-21950-6	1104		X	JT8D9A *[2]		P	P
65-22249-3	1105	X	X	JT8D17A		P	P
65-22454-8	1106	X	X				
65-45798-2	1109	X	X				
65-45798-4	1109	X	X				
65-45798-10	1109	X	X				
65-45799-171	1109	X	X	*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C			
65-46851-5 *[1]	1111	X	X				
65-51534-5	1110	X	X	*[2] ENGINE INTERMIX PER MC 3510-221			
65-52971-1	1117	X	X				
65-54476-4 *[1]	1118	X	X	*[3] ENGINE INTERMIX PER SERVICE BULLETIN 71-1337 AND MC 7200MK3158			
65-55749-9 *[1]	1120	X	X				
65-58122-2 *[1]	1123	X	X				
65-58125-3	1124	X	X				
65-58126-20	1125	X	X				
65-58126-24	1125	X	X				
65-58135-21	1126	X					
65-58135-47 *[1]	1126	X	X				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-1	1142A	X	X				
65-70950-7	1138	X					
65-70950-8	1138		X				
65-77451-4	1125	X	X				
65-85379-3	1143	X					
65-85379-4	1143		X				
69-15863-1	1130	X	X				
69-48936-1	1137	X	X				
JT8D15ABG300		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (VM)

AIRLINE (VP)				AIRLINE (VP)			
MODEL 2A1C				MODEL 214			
ENGINE 1 65-73769-23				ENGINE 1 65-73769-37			
ENGINE 2 65-73769-24				ENGINE 2 65-73769-38			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-10	1103	X	X	65-19684-1	1102	X	X
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-2	1105	X	X
65-22249-3	1105	X	X	65-22454-13	1106	X	X
65-45799-148	1109	X	X	65-45798-1	1109	X	X
65-46851-4	1111	X	X	65-46853-1	1113	X	X
65-51534-1	1110	X	X	65-51534-1	1110	X	X
65-52971-1	1117	X	X	65-52261-1	1116	X	X
65-55475-28	1119	X	X	65-52971-1	1117	X	X
65-58125-3	1124	X	X	65-55750-1	1120	X	
65-58135-18	1126	X		65-55750-2	1120		X
65-58135-20	1126		X	65-58125-1	1124	X	X
65-58135-26	1126	X	X	65-58135-18	1126	X	
65-58145-12	1127	X	X	65-58135-20	1126		X
65-58234-10	1128	X	X	65-58145-1	1127	X	X
65-58234-3	1128	X	X	65-58145-4	1127	X	X
65-65919-4	1108	X	X	65-58234-10	1128	X	X
65-69659-1	1142A	X	X	65-58234-3	1128	X	X
65-70950-11	1138	X		65-65919-4	1108	X	X
65-70950-12	1138		X	65-69659-2	1142A	X	X
65-85379-3	1143	X		65-70950-1	1138	X	
65-85379-4	1143		X	65-70950-2	1138		X
69-28727-8	1133	X	X	69-28727-1	1133	X	X
69-48936-1	1137	X	X	69-48936-1	1137	X	X
JT8D17BG42		X	X	JT8D7BG3		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

AIRLINE (VP)
MODEL 2A1, 2N3
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-69659-1	1142A	X	X
65-20996-10	1103	P	P	65-70950-1	1138	P	
65-21950-3	1104	P	P	65-70950-2	1138		P
65-21950-5	1104	P		65-70950-11	1138	P	
65-21950-6	1104		P	65-70950-12	1138		P
65-22249-3	1105	X	X	65-70950-5	1138	P	
65-22454-8	1106	X	X	65-70950-6	1138		P
65-45798-1	1109	P	P	65-77451-2	1125	P	P
65-45798-2	1109	P	P	65-77451-3	1125	P	P
65-45798-5	1109	P	P	65-85379-1	1143	P	
65-45799-127	1109	P	P	65-85379-2	1143		P
65-45799-147	1109	P	P	65-85379-3	1143	P	
65-45799-148	1109	P	P	65-85379-4	1143		P
65-46851-3	1111	P	P	69-15863-1	1130	P	P
65-46851-4	1111	P	P	69-28727-8	1133	P	P
65-46851-5	1111	P	P	69-48936-1	1137	X	X
65-46853-1	1113	P	P	JT8D17BG42		P	P
65-51534-1	1110	X	X	JT8D9BG21		P	P
65-52261-1	1116	P	P				
65-52971-1	1117	X	X				
65-55475-28	1119	P	P				
65-55750-1	1121	P					
65-55750-2	1121		P				
65-58125-3	1124	X	X				
65-58126-20	1125	X	X				
65-58135-18	1126	X					
65-58135-20	1126		X				
65-58135-21	1126	X					
65-58135-26	1126	P	P				
65-58145-12	1127	P	P				
65-58145-4	1127	P	P				
65-58234-10	1128	X	X				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (VP)
 MODEL 2A1
 ENGINE 1 65-73769-65
 ENGINE 2 65-73769-66

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19684-1	1102	X	X
65-21950-3	1104	X	X
65-22249-3	1105	X	X
65-22454-8	1106	X	X
65-45798-1	1109	X	X
65-45799-147	1109	P	P
65-45799-148	1109	P	P
65-46853-1	1113	X	X
65-51534-1	1110	X	X
65-52261-1	1116	X	X
65-52971-1	1117	X	X
65-55750-1	1121		X
65-55750-2	1121		X
65-58125-1	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-1	1127	X	X
65-58145-4	1127	X	X
65-58234-10	1128	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-1	1138	P	
65-70950-2	1138		P
65-70950-11	1138	P	
65-70950-12	1138		P
65-28727-1	1133	X	X
65-48936-1	1137	X	X
JT8D9BG3		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (VZ)				AIRLINE (WA)			
MODEL 2N1				MODEL 247			
ENGINE 1 65-73769-51				ENGINE 1 65-73769-7			
ENGINE 2 65-73769-52				ENGINE 2 65-73769-8			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-20996-10	1103	X	X	65-19684-1	1102	X	X
65-21950-5	1104	X		65-21950-3	1104	X	X
65-21950-6	1104		X	65-22249-3	1105	X	X
65-22249-3	1105	X	X	65-22454-3	1106	P	P
65-22454-8	1106	X	X	65-22454-7	1106	P	P
65-45798-5	1109	X	X	65-27808-1	1108	P	P
65-45799-147	1109	P	P	65-45798-1	1109	X	X
65-45799-148	1109	P	P	65-46853-1	1113	X	X
65-46851-5	1111	X	X	65-50882-2	1115	X	X
65-52971-1	1117	X	X	65-51534-1	1110	X	X
65-55475-28	1119	X	X	65-52261-1	1116	X	X
65-58125-3	1124	X	X	65-52971-1	1117	X	X
65-58126-20	1125	X	X	65-55475-21	1119	X	X
65-58135-18	1126	X		65-55750-1	1120	X	
65-58135-20	1126		X	65-55750-2	1120		X
65-58135-21	1126	X		65-57815-1	1140	P	P
65-58135-26	1126	X	X	65-57815-2	1140	P	P
65-58145-12	1127	X	X	65-58125-1	1124	X	X
65-58134-10	1128	X	X	65-58135-12	1126	P	P
65-58134-3	1128	X	X	65-58135-18	1126	P	
65-65919-4	1108	X	X	65-58135-20	1126		P
65-70950-11	1138	X		65-58145-4	1127	X	X
65-70950-12	1138		X	65-58145-9	1127	P	P
65-70950-5	1138	X		65-58234-10	1128	P	P
65-70950-6	1138		X	65-58234-3	1128	X	X
65-77451-3	1125	X	X	65-65919-4	1108	P	P
65-85379-3	1140	X		65-69659-7	1142A	X	X
65-85379-4	1140		X	65-70950-1	1138	P	
69-28727-8	1133	X	X	65-70950-2	1138		P
69-48936-1	1137	X	X	65-70950-5	1138	P	
JT8D17BG42		X	X	65-70950-6	1138		P
				65-73778-1	1138	P	P
				69-15863-1	1130	X	X
				69-21612-1	1132	X	X
				69-48936-1	1137	X	X
				JT8D9BG4		P	P
				JT8D9A *[1]		P	P
				JT8D15 *[1]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] ENGINE INTERMIX PER MC 3510-85K AND SB 71-1162

BOEING 
COMMERCIAL JET
OVERHAUL MANUAL

737
POWERPLANT BUILDUP

AIRLINE (WA)
MODEL 2T2
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23	*[1] 1101	X		65-70950-11	1138	X	
65-19592-24	*[1] 1101		X	65-70950-12	1138		X
65-20996-1	*[2] 1103	X	X	65-77451-44	1125	X	X
65-21950-5	1104	X		65-85379-3	1143	X	
65-21950-6	1104		X	65-85379-4	1143		X
65-22249-1	1105	X	X	65C13071-3	1130	X	X
65-22454-8	1106	X	X	69-28727-8	1133	X	X
65-45798-5	*[1] 1109	X	X	69-48936-1	1137	X	X
65-45798-10	*[1] 1109	X	X	69-56334-1	1139	X	X
65-45799-147	1109	X	X	JT8D17BG304		X	X
65-45799-148	1109	X	X				
65-46851-5	*[1] 1111	X	X				
65-51534-5	1110	X	X				
65-52971-1	1117	X	X				
65-54476-4	*[1] 1118	X	X				
65-55475-28	*[1] 1119	X	X				
65-55749-9	*[1] 1120	X	X				
65-58122-2	*[1] 1123	X	X				
65-58125-3	1124	X	X				
65-58126-20	1125	X	X				
65-58135-20	1126		X				
65-58135-21	1126A	X					
65-58135-30	*[1] 1126	X	X				
65-58135-47	*[1] 1126	X	X				
65-58145-15	*[1] 1127	X	X				
65-58234-3	1128	X	X				
65-65919-4	1108	X	X				
65-69659-2	1142A	X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1007

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP SYMBOLS

SYMBOLS X= USED ON ALL ENGINES FOR POSITION SHOWN
P= PARTIAL USAGE

OVERHAUL MANUAL

AIRLINE (WA)
MODEL 2T4
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-19592-23	*[1] 1101	X	
65-19592-24	*[1] 1101		X
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-2	1105	X	X
65-45798-10	*[1] 1109	X	X
65-45799-148	1109	X	X
65-46851-4	*[1] 1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-54476-4	*[1] 1118	X	X
65-55749-9	*[1] 1120	X	X
65-58122-2	*[1] 1123	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58135-30	*[1] 1126	X	X
65-58135-47	*[1] 1126	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-2	1142A	X	X
65-70950-11	1138	X	
65-70950-12	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D17BG301		X	X

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X= USED ON ALL ENGINES FOR POSITION SHOWN
P= PARTIAL USAGE

AIRLINE (WA)
MODEL 247
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (WA)
MODEL 2J8
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-19592-23 *[1]	1101	X	
65-19592-24 *[1]	1101		X	65-19592-24 *[1]	1101		X
65-20996-21 *[1]	1103	X	X	65-21950-5	1104	X	
65-21931-23	1114A	X	X	65-21950-6	1104		X
65-21950-5	1104	X		65-22249-3	1105	X	X
65-21950-6	1104		X	65-45798-4 *[1]	1109	X	X
65-22249-2	1105	X	X	65-45798-6	1109	X	X
65-45798-4 *[1]	1109	X	X	65-45798-10	1109	X	X
65-45798-6	1109	X	X	65-45799-148	1109	X	X
65-45798-7	1109	X	X	65-46851-4 *[1]	1111	X	X
65-45798-10	1109	X	X	65-51534-8	1110	X	X
65-45799-165	1109	X	X	65-52971-1	1117	X	X
65-45799-170	1109	X	X	65-54476-4 *[1]	1118	X	X
65-46851-4 *[1]	1111	X	X	65-55749-9 *[1]	1120	X	X
65-51534-8	1110	X	X	65-58122-2 *[1]	1123	X	X
65-54476-4 *[1]	1118	X	X	65-58125-3	1124	X	X
65-55475-33 *[1]	1119	X	X	65-58135-18	1126	X	
65-55749-8 *[1]	1120	X	X	65-58135-20	1126		X
65-56646-3 *[1]	1122	X	X	65-58135-30 *[1]	1126		X
65-58122-2 *[1]	1123	X	X	65-58135-47 *[1]	1126		X
65-58125-3	1124	X	X	65-58234-3	1128	X	X
65-58135-18	1126	X		65-65919-4	1108	X	X
65-58135-20	1126		X	65-69659-1	1142A	X	X
65-58135-47 *[1]	1126	X	X	65-70950-11	1138	X	
65-58234-20	1128	X	X	65-70950-12	1138		X
65-65919-4	1108	X	X	65-85379-3	1143	X	
65-69659-1	1142A	X	X	65-85379-4	1143		X
65-70950-7	1138	X		69-15863-1	1130	X	X
65-70950-8	1138		X	69-48936-1	1137	X	X
65-85379-3	1143	X		69-50911-2 *[1]	1141	X	X
65-85379-4	1143		X	JT8D17BG301 *[1]		P	P
69-15863-1	1130	X	X	JT8D9 *[1]		P	P
69-48936-1	1137	X	X	JT8D9A *[1]		P	P
69-50911-2 *[1]	1141	X	X	JT8D15 *[2]		P	P
JT8D15ABG304 *[3]		P	P	JT8D15A *[2]		P	P
JT8D-9 *[3]		P	P				
JT8D-9A *[3]		P	P				
JT8D-15A *[4]		P	P				
JT8D-17A *[4]		P	P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

*[2] ENGINE INTERMIX PER RR 97012-3

*[3] ENGINE INTERMIX PER MC 3510-219K AND SB 71-1156

*[4] ENGINE INTERMIX PER MC 3510-215K

POWER PACK INSTALLATIONS

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (WE)

MODEL 210

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-3	1101	X	X	65-58135-18	1126	X	
65-20966-1	1103	X	X	65-58135-20	1126		X
65-21950-5	1104	X		65-58135-30	1126	X	X
65-21950-6	1104		X	65-58145-10	1127	X	X
65-22249-2	1105	X	X	65-58234-3	1127	X	X
65-25736-2	1107	X	X	65-65919-4	1108	X	X
65-45798-3	1109	X	X	65-69659-1	1142A	X	X
65-45798-5	1109	X	X	65-70950-11	1138	X	
65-46851-1	1111	X	X	65-70950-12	1138		X
65-46851-5	1111	X	X	65-77451-4	1125	X	X
65-46852-1	1112	X	X	JT8D17BG301		X	X
65-46853-2	1113	X	X				
65-50571-1	1114	X	X				
65-50882-1	1115	X	X				
65-51534-1	1110	X	X				
65-52261-2	1116	X	X				
65-52971-1	1117	X	X				
65-54476-1	1118	X	X				
65-55475-1	1119	X	X				
65-55749-1	1120	X	X				
65-56646-1	1122	X	X				
65-58122-1	1123	X	X				
65-58125-3	1124	X	X				
65-58126-1	1125	X	X				
65-58126-2	1125	X	X				
65-58126-20	1125	X	X				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE
 BUILDUP SYMBOLS X= USED ON ALL ENGINES FOR POSITION SHOWN
 P= PARTIAL USAGE

737

POWERPLANT BUILDUP



OVERHAUL MANUAL

AIRLINE (WE)
 MODEL 202C, 210C
 ENGINE 1 65-73769-23
 ENGINE 2 65-73769-24

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	65-69659-2	1142A	P	P
65-20996-1*[2]	1103	X	X	65-70950-1	1138	P	
65-21950-3	1104	X	X	65-70950-2	1138		P
65-21950-5	1104	X	X	65-70950-5	1138	P	
65-22249-2	1105	X	X	65-70950-6	1138		P
65-22454-8	1106	X	X	65-70950-11	1138	P	
65-27808-1	1108	P	P	65-70950-12	1138		P
65-45798-1	1109	P	P	65-73778-1 (SB 78-1005)		P	P
65-45798-2	1109	P	P	65-77451-3	1125	P	P
65-45798-5	1109	P	P	65-77451-4	1125	P	P
65-45799-127	1109	X	X	65-85379-3	1143	P	
65-46851-3	1111	P	P	65-85379-4	1143		P
65-46851-5	1111	P	P	69-15863-1	1130	X	X
65-46853-1	1113	P	P	69-21612-1	1132	P	P
65-51534-1	1110	P	P	69-48936-1	1137	X	X
65-51534-5	1110	P	P	JT8D9ABG42 *[1]		P	P
65-52261-1	1116	P	P	JT8D9BG3 *[1]		P	P
65-52971-1	1117	X	X	JT8D9BG42 *[1]		P	P
65-55750-1	1120	P		JT8D9BG9 *[1]		P	P
65-55750-2	1120		P	JT8D17BG301 *[1]		P	P
65-57815-1	1140	P	P	JT8D17BG42 (OPT TO		P	P
65-57815-2	1140	P	P	JT8D17BG301) *[1]			
65-58125-1	1124	P	P				
65-58125-3	1124	P	P				
65-58126-20	1125	P	P				
65-58135-12	1126	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58135-21	1126	P					
65-58135-30	1126	P	P				
65-58145-4	1127	P	P				
65-58234-10	1128	P	P				
65-58234-3	1128	X	X				
65-65919-4	1108	P	P				
65-69659-1	1142A	P	P				

*[1] ENGINE INTERMIX JT8D-9/-9A WITH
 JT8D-17 PER MC 3510-8K AND
 SB 71-1060

*[2] INCLUDES ENGINE STARTER VALVE
 OPEN INDICATION LIGHT PRESSURE
 SWITCH PER MC 3540-11K AND
 SB 80-1013

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

POWER PACK INSTALLATIONS
 (WE)

Dec 5/84

71-03-01
 Page 1285

OVERHAUL MANUAL

AIRLINE (WT)
MODEL 2F9
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

AIRLINE (WT)
MODEL 2F9
ENGINE 1 65-73769-51
ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	X	X	69-19592-23 *[1]	1101	X	
65-20996-1 *[2]	1103	X	X	65-19592-24 *[1]	1101		X
65-21950-3	1104	X	X	65-20996-1 *[2]	1103	X	X
65-22249-1	1105	X	X	65-21950-5	1104	X	
65-22454-8	1106	X	X	65-21950-6	1104		X
65-45799-134	1109	X	X	65-22249-1	1105	X	X
65-46851-4	1111	X	X	65-45798-3 *[1]	1109	P	P
65-51534-1	1110	X	X	65-45798-8 *[1]	1109	P	P
65-52971-1	1117	X	X	65-45798-10 *[1]	1109	P	P
65-55475-26	1119	X	X	65-45799-1	1109	P	P
65-55750-1	1120	X		65-45799-170	1109	P	P
65-55750-2	1120		X	65-46851-4 *[1]	1111	X	X
65-58125-3	1124	X	X	65-51534-5	1110	X	X
65-58135-18	1126	X		65-52971-1	1117	X	X
65-58135-20	1126		X	65-54476-1 *[1]	1118	P	P
65-58145-1	1127	X	X	65-54476-4 *[1]	1118	P	P
65-58234-10	1128	X	X	65-55475-28 *[1]	1119	X	X
65-58234-3	1128	X	X	65-55749-1 *[1]	1120	P	P
65-65919-4	1108	X	X	65-55749-9 *[1]	1120	P	P
65-69659-1	1142A	X	X	65-58122-1 *[1]	1123	P	P
65-70950-3	1138	X		65-58122-2 *[1]	1123	P	P
65-70950-4	1138		X	65-58125-3	1124	X	X
69-28727-1	1133	X	X	65-58135-4 *[1]	1126	P	P
69-48936-1	1137	X	X	65-58135-47 *[1]	1126	P	P
JT8D15BG38 *[3]		P	P	65-58135-18	1126	X	
JT8D15A *[3]		P	P	65-58135-20	1126		X
				65-58145-12 *[1]	1127	X	X
*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C				65-58234-3	1128	P	P
*[2] INCLUDES ENGINE STARTER VALVE OPEN INDICATION LIGHT PRESSURE SWITCH PER MC 3540-11K AND SB 80-1010				65-58234-16	1128	P	P
*[3] ENGINE INTERMIX PER MC 3510-137K, -141K AND SB 71-1097, 71-1098				65-65919-4	1108	X	X
				65-69659-1	1142A	P	P
				65-69659-2	1142A	P	P
				65-70950-7	1138	X	
				65-70950-8	1138		X
				65-85379-3	1143	X	
				65-85379-4	1143		X
				69-28727-8	1133	X	X
				69-48936-1	1137	X	X
				JT8D15ABG310 *[3]		P	P
				JT8D15BG310 *[3]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

737

POWERPLANT BUILDUP



AIRLINE (WU)
 MODEL 2S3
 ENGINE 1 65-73769-51
 ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE	
		1	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-46851-4	1111	X	X
65-51534-5	1110	X	X
65-52971-1	1117	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-65919-4	1108	X	X
65-69659	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1140	X	
65-85379-4	1140		X
69-15863-1	1130	X	X
69-48936-1	1137	X	X
JT8D15BG301		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP
 SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN

AIRLINE (XIA)

MODEL 25C

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-29 *[1]	1101	X		65-58122-2 *[1]	1123	X	X
65-19592-30 *[1]	1101		X	65-58125-3	1124	X	X
65-20996-24 *[1]	1103	X	X	65-58126-1	1125	X	X
65-21931-23	1114A	X	X	65-58126-2	1125	X	X
65-21950-5	1104	X		65-58126-24	1125	X	X
65-21950-6	1104		X	65-58135-1	1126	X	X
65-22249-2	1105	X	X	65-58135-2	1126	X	X
65-25736-2	1107	X	X	65-58135-3	1126	X	X
65-45798-4	1109	X	X	65-58135-18	1126	X	
65-45798-7	1109	X	X	65-58135-20	1126		X
65-45798-10	1109	X	X	65-58135-47 *[1]	1126	X	X
65-45799-148	1109	X	X	65-58145-10	1127	X	X
65-45799-166	1109	X	X	65-58234-19	1128	X	X
65-46851-4 *[1]	1111	X	X	65-69659-1	1142A	X	X
65-46852-1	1112	X	X	65-70950-7	1138	X	
65-46853-2	1113	X	X	65-70950-8	1138		X
65-46858-131	1138	X		65-85379-3	1143	X	
65-46858-132	1138		X	65-85379-4	1143		X
65-50882-1	1115	X	X	69-15863-1	1130	X	X
65-51534-5	1110	X	X	69-15935-1	1131	X	X
65-52261-2	1116	X	X	69-36670-1	1134	X	X
65-52971-1	1117	X	X	69-37478-1	1135	X	X
65-54476-4 *[1]	1118	X	X	69-45172-1	1136	X	X
65-55475-33 *[1]	1119	X	X	69-48936-1	1137	X	X
65-55749-8 *[1]	1120	X	X	69-50911-2 *[1]	1141	X	X
65-56646-3 *[1]	1122	X	X	69-60143-1	1142	X	X
				JT8D15ABG300			

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

737

POWERPLANT BUILDUP

BOEING 
COMMERCIAL JET
 OVERHAUL MANUAL

AIRLINE (YB)

MODEL 2Y5

ENGINE 1 65-73769-51

ENGINE 2 65-73769-52

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19592-23 *[1]	1101	X		65-58126-1	1125	X	X
65-19592-24 *[1]	1101		X	65-58126-2	1125	X	X
65-20996-19 *[1]	1103	P	P	65-58126-24	1125	P	P
65-20996-23 *[1]	1103	P	P	65-68135-1	1126	X	X
65-21950-5	1104	X		65-58135-2	1126	X	X
65-21950-6	1104		X	65-58135-3	1126	X	X
65-22249-1	1105	X	X	65-58135-18	1126	X	
65-25736-3	1107	X	X	65-58135-20	1126		X
65-45798-3	1109	X	X	65-58135-47 *[1]	1126	X	X
65-45798-4	1109	P	P	65-58145-12 *[1]	1127	X	X
65-45798-8	1109	P	P	65-58234-3	1128	P	P
65-45798-10	1109	X	X	65-58234-16	1128	P	P
65-45799-170	1109	X	X	65-65919-4	1108	X	X
65-46851-4 *[1]	1111	X	X	65-69659-1	1142A	X	X
65-46852-1	1112	X	X	65-70950-7	1138	X	
65-46853-2	1113	X	X	65-70950-8	1138		X
65-46858-131	1138	X		65-85379-3	1143	X	
65-46858-132	1138		X	65-85379-4	1143		X
65-50571-1	1114	P	P	69-15935-1	1131	X	X
65-50571-7	1114	P	P	69-21612-1	1132	X	X
65-50882-1	1115	X	X	69-28727-8	1133	X	X
65-51534-5	1110	X	X	69-36670-1	1134	X	X
65-52261-2	1116	X	X	69-37478-1	1135	X	X
65-52971-1	1117	X	X	69-45172-1	1136	X	X
65-54476-4 *[1]	1118	X	X	69-48936-1	1137	X	X
65-55475-28 *[1]	1119	X	X	69-50911-1	1141	X	X
65-55749-9 *[1]	1120	X	X	69-56334-1	1139	X	X
65-56646-1	1122	X	X	JT8D15ABG314		P	P
65-58122-2 *[1]	1123	P	P	JT8D15ABG300		P	P
65-58122-3 *[1]	1123	P	P				
65-58125-3	1124	X	X				

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C

INSTALLATION LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

737
POWERPLANT BUILDUP

OVERHAUL MANUAL

AIRLINE (ZD)
MODEL 204, 204C
ENGINE 1 65-73769-31
ENGINE 2 65-73769-32

INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-19684-1	1102	P	P	69-15863-1	1130	P	P
65-21950-3	1104	X	X	69-21612-1	1132	X	X
65-22249-3	1105	X	X	69-28727-8	1133	P	P
65-22454-9	1105	X	X	69-48936-1	1137	X	X
65-27808-1	1108	P	P	JT8D15BG44		P	P
65-45798-1	1109	P	P	JT8D9BG22		P	P
65-46851-4	1111	P	P	JT8D9BG3		P	P
65-46853-1	1113	P	P	JT8D9BG9		P	P
65-51534-1	1110	P	P				
65-51534-5	1110	P	P				
65-52261-1	1116	P	P				
65-52971-1	1117	X	X				
65-55475-28	1119	P	P				
65-55750-1	1120	P					
65-55750-2	1120		P				
65-58125-1	1124	P	P				
65-58125-3	1124	P	P				
65-58135-12	1126	P	P				
65-58135-18	1126	P					
65-58135-20	1126		P				
65-58145-12	1127	P	P				
65-58145-4	1127	P	P				
65-58234-10	1128	P	P				
65-58234-3	1128	X	X				
65-65919-4	1108	P	P				
65-69659-1	1142A	P	P				
65-69659-7	1142A	P	P				
65-70950-1	1138	P					
65-70950-2	1138		P				
65-70950-5	1138	P					
65-70950-6	1138		P				
65-70950-7	1138	P					
65-70950-8	1138		P				
65-85379-3	1143	P					
65-85379-4	1143		P				

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

AIRLINE (ZD)				AIRLINE (ZD)			
MODEL 204, 2U4				MODEL 204			
ENGINE 1 65-73769-31				ENGINE 1 65-73769-31			
ENGINE 2 65-73769-32				ENGINE 2 65-73769-32			
INSTALLATION	FIGURE	ENGINE		INSTALLATION	FIGURE	ENGINE	
		1	2			1	2
65-21950-5	1104	X		65-19592-23 *[1]	1101	X	
65-21950-6	1104		X	65-19592-24 *[1]	1101		X
65-22249-1	1105	X	X	65-21950-5	1104	X	
65-22454-8	1106	X	X	65-21950-6	1104		X
65-45799-1	1109	X	X	65-22249-3	1105	X	X
65-46851-4	1111	X	X	65-45798-8	1109	X	X
65-51534-5	1110	X	X	65-45799-170	1109	X	X
65-55475-28	1119	X	X	65-46851-4 *[1]	1111	X	X
65-58125-3	1124	X	X	65-51534-5	1110	X	X
65-58135-18	1126	X		65-52971-1	1117	X	X
65-58135-20	1126		X	65-55475-28 *[1]	1119	X	X
65-58145-12	1127	X	X	65-58125-3	1124	X	X
65-58234-3	1128	X	X	65-58135-18	1126	X	
65-65919-4	1108	X	X	65-58135-20	1126		X
65-69659-1	1142A	X	X	65-58145-12 *[1]	1127	X	X
65-69659-2	1142A	X	X	65-58234-16	1128	X	X
65-70950-7	1136	X		65-65919-4	1108	X	X
65-70950-8	1136		X	65-69659-1	1142A	X	X
65-85379-3	1143	X		65-69659-2	1142A	X	X
65-85379-4	1143		X	65-70950-7	1138	X	
69-21612-1	1132	X	X	65-70950-8	1138		X
69-28727-8	1133	X	X	65-85379-3	1143	X	
69-48936-1	1136	X	X	65-85379-4	1143		X
JT8D15BG301 *[2]		P	P	69-21612-1	1132	X	X
JT8D17 *[2]		P	P	69-28727-8	1133	X	X
JT8D15 *[3]*[4]		P	P	69-48936-1	1137	X	X
JT8D17A *[3]		P	P				
JT8D9A *[4]		P	P	JT8D15ABG301 *[2]		P	P
				JT18D17 *[2]		P	P
				JT8D15 *[3]*[4]		P	P
				JT8D17A *[3]		P	P
				JT8D9A *[4]		P	P

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS X = USED ON ALL ENGINES FOR POSITION SHOWN
P = PARTIAL USAGE

- *[1] SUPERSEDES INSTALLATION LISTED IN TABLE C
- *[2] ENGINE INTERMIX PER MC 3510-229K AND SB 71-1155
- *[3] ENGINE INTERMIX PER MC 3530-217K AND SB 71-1159
- *[4] ENGINE INTERMIX PER MC 3510-230K AND SB 71-1161

AIRLINE (ZD)
 MODEL 204
 ENGINE 1 65-73769-31
 ENGINE 2 65-73769-32

INSTALLATION	FIGURE	ENGINE	
		I	2
65-21950-5	1104	X	
65-21950-6	1104		X
65-22249-3	1105	X	X
65-45799-1	1109	X	X
65-46851-4 *[1]	1111	X	X
65-51534-1	1110	X	X
65-52971-1	1117	X	X
65-55475-28 *[1]	1119	X	X
65-58125-3	1124	X	X
65-58135-18	1126	X	
65-58135-20	1126		X
65-58145-12 *[1]	1127	X	X
65-58234-3	1128	X	X
65-65919-4	1108	X	X
65-69659-1	1142A	X	X
65-70950-7	1138	X	
65-70950-8	1138		X
65-85379-3	1143	X	
65-85379-4	1143		X
69-21612-1	1132	X	X
69-28727-8	1133	X	X
69-48936-1	1137	X	X
JT8D15BG42		X	X

INSTALLATIONS LISTED IN TABLE C MUST ALSO BE INCLUDED FOR COMPLETE ENGINE BUILDUP

SYMBOLS: X = USED ON ALL ENGINES FOR POSITION SHOWN
 P = PARTIAL USAGE

*[1] SUPERSEDES INSTALLATION LISTED IN TABLE C