

Three-Dimensional Vibrations of Tethered Satellite Systems

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The three-dimensional free vibrations of an orbiter-towed tethered satellite system with nominally vertical tether during stationkeeping phase are studied analytically. The tether is modeled as an elastic continuum with mass and the orbiter and the subsatellite as two concentrated masses. The linearized equations of motion with variable coefficients are solved by means of a perturbation technique. An analysis of the order of magnitude of the different terms involved permits evaluation of the coupling between longitudinal and transversal in-plane vibrations coming to an approximate closed-form solution of the eigenvalue problem.

Introduction

THE vast potential of the tethered satellite system (TSS)^{1,2} within the space research program has recently stimulated many investigations on its dynamic behavior and control. Most of the work deals with deployment and retrieval of the subsatellite and the determination of the most appropriate control law.³⁻⁵ Yet despite its importance, little attention has been devoted to the dynamics of the stationkeeping phase.⁶⁻¹⁰ Knowledge of the elastic oscillations of the system is therefore necessary to evaluate the disturbances introduced into the experimental observations. A survey on the development in these areas has been provided by Misra and Modi.¹¹

A continuum model was formulated previously¹² by the authors in order to investigate the free dynamics of the TSS during the stationkeeping phase. The following simplifying assumptions were introduced: 1) the orbiter and the satellite are considered as two concentrated masses, 2) the tether is modeled as an elastic continuum with mass, and 3) the orbital eccentricity and the aerodynamic drag forces are neglected so that the tether can be assumed vertical. The nonlinear equations of motion have been derived through an energy method and then linearized around the static configuration. Out-of-plane oscillations have been studied separately because the relevant linear equations are uncoupled from in-plane equations of motion.

In this paper, by using the model previously introduced, the in-plane motion is analyzed. The governing equations in the longitudinal and transversal components are coupled by means of gyroscopic forces, leading to a nonstandard eigenvalue problem.^{13,14} It is found that the eigenvalue problem in the real and imaginary parts of the eigenfunctions of the gyroscopic system splits into two equal problems because of the particular structure of the mass and stiffness operators. A perturbation technique is used to solve the system characterized by variable coefficients arising from the stress gradient along the tether. As a result, eigenvalues and eigenfunctions are determined by perturbing those relative to a constant state of stress. In order to evaluate the influence of different terms in the equations of motion, an asymptotic analysis has been

performed. This shows that, within technical values of the geometric and mechanical parameters, the two components of motion are weakly coupled by gyroscopic forces and two sets of eigenfunctions exist, one prevalently longitudinal and the other prevalently transversal. These conclusions are supported by numerical results.

In the authors' opinion, some new contributions have been made with respect to previous papers, arising from 1) the use of a more sophisticated model and 2) the use of perturbation techniques in computing eigenfrequencies and eigenvectors. In particular, the gyroscopic forces associated with the in-plane motion are taken into account and no assumptions are introduced on the value of the satellite/orbiter mass ratio. As a consequence, the model permits a more accurate description of the motion and enables the systems to be analyzed with lumped masses of the same order. Furthermore, the perturbation method employed permits attainment of a simple closed-form solution, in comparison, for example, with Ref. 10, although it cannot always be applied. Finally, the asymptotic analysis based on the estimate of the order of magnitude of the terms of the equations of motion provides important insights into the mechanical behavior of the system.

Formulation of the Problem

Let us consider the system consisting of a Space Shuttle (SS) and a subsatellite, connected by an elastic tether. Let (O, X, Y, Z) be a rest frame whose origin coincides with the Earth mass center. Assume that the SS describes a circular, undisturbed orbit of radius a in the X - Y plane, with constant angular velocity n . It is convenient to introduce a reference system (o, x, y, z) whose origin coincides with the SS center of mass; the x, y axes lie in the X - Y plane and are oriented as shown in Fig. 1. A curvilinear abscissa S measured from the SS along the tether is also introduced.

Accounting for the kinetic and potential energies, the Lagrangian of the system is

$$\begin{aligned}
 L = & \frac{1}{2} \left\{ \mu \int_0^l |\dot{\mathbf{r}} + \mathbf{n} \times \mathbf{r}|^2 dS + m_1 |\dot{\mathbf{r}} + \mathbf{n} \times \mathbf{r}|_{S=0}^2 \right. \\
 & \left. + m_2 |\dot{\mathbf{r}} + \mathbf{n} \times \mathbf{r}|_{S=l}^2 \right\} \\
 & + \mu_e \left\{ \mu \int_0^l dS / |\mathbf{r}| + m_1 / |\mathbf{r}|_{S=0} + m_2 / |\mathbf{r}|_{S=l} \right\} \\
 & - \frac{1}{2} E A \int_0^l e^2 dS
 \end{aligned} \quad (1)$$

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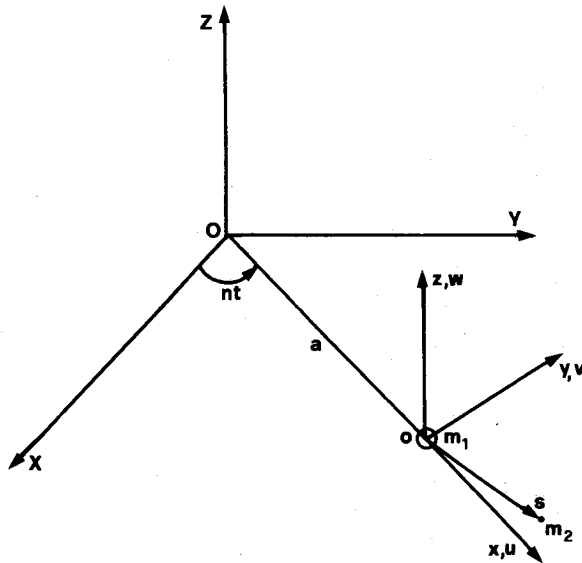


Fig. 1 TSS model.

where a dot indicates differentiation with respect to time \bar{t} . In Eq. (1), μ_e denotes the gravitational constant, μ the mass density of the tether, ℓ its unstretched length, A its cross section, E the Young's modulus, and m_1 and m_2 the masses of the SS and the subsatellite, respectively. In addition, r is the position vector, $r = (a+x)c_1 + y c_2 + z c_3$, where c_1, c_2, c_3 are the unit vectors of the moving system, and e is the stretching defined as $e = [(x'^2 + y'^2 + z'^2)^{1/2} - 1]$, where a prime denotes differentiation with respect to S .

From the stationary condition of the Lagrangian, the general equations of motion are obtained

$$\begin{aligned} \mu \ddot{x} - 2\mu n \dot{y} - \mu n^2(a+x) \\ + \mu \mu_e \frac{(a+x)}{|r|^3} - E A \left[x' \frac{e}{(1+e)} \right]' = 0 \end{aligned} \quad (2a)$$

$$\begin{aligned} \mu \ddot{y} + 2\mu n \dot{x} - \mu n^2 y + \mu \mu_e \frac{y}{|r|^3} \\ - E A \left[y' \frac{e}{(1+e)} \right]' = 0 \end{aligned} \quad (2b)$$

$$\mu \ddot{z} + \mu \mu_e \frac{z}{|r|^3} - E A \left[z' \frac{e}{(1+e)} \right]' = 0 \quad (2c)$$

with appropriate boundary conditions.

Equations (2) admit a steady-state solution $x = x(S)$, $y = 0$, $z = 0$, where $x(S)$ is obtained by solving the nonlinear equation

$$-\mu n^2(a+x) + \frac{\mu \mu_e}{(a+x)^2} - E A x'' = 0 \quad (3)$$

with relevant boundary conditions.

By considering the motion with respect to the static configuration and linearizing in the additional displacement components U, V, W , along x, y, z axes, respectively, we get the nondimensional equations of motion

$$\alpha^2 \ddot{u} - 2\alpha^2 \dot{v} - 3\alpha^2 u - u'' = 0 \quad (4a)$$

$$\alpha^2 \ddot{v} + 2\alpha^2 \dot{u} - \alpha^2 [f(s)v']' = 0 \quad (4b)$$

$$\alpha^2 \ddot{w} + \alpha^2 w - \alpha^2 [f(s)w']' = 0 \quad (4c)$$

with the associated boundary conditions

$$[\alpha^2 \ddot{u} - 2\alpha^2 \dot{v} - 3\alpha^2 u - \beta u']_{s=0} = 0 \quad (5a)$$

$$[\alpha^2 \ddot{u} - 2\alpha^2 \dot{v} - 3\alpha^2 u + \gamma u']_{s=1} = 0 \quad (5b)$$

$$[\alpha^2 \ddot{v} + 2\alpha^2 \dot{u} - \alpha^2 \beta f(s)v']_{s=0} = 0 \quad (5c)$$

$$[\alpha^2 \ddot{v} + 2\alpha^2 \dot{u} + \alpha^2 \gamma f(s)v']_{s=1} = 0 \quad (5d)$$

$$[\alpha^2 \ddot{w} + \alpha^2 w - \alpha^2 \beta f(s)w']_{s=0} = 0 \quad (5e)$$

$$[\alpha^2 \ddot{w} + \alpha^2 w + \alpha^2 \gamma f(s)w']_{s=1} = 0 \quad (5f)$$

where $u = U/\ell$, $v = V/\ell$, $w = W/\ell$, and a prime and a dot denote differentiations with respect to $s = S/\ell$ and $t = \bar{t}/n$. The following nondimensional parameters have also been introduced

$$\alpha^2 = \frac{\mu n^2 \ell^2}{E A}, \quad \beta = \frac{\mu \ell}{m_1}, \quad \gamma = \frac{\mu \ell}{m_2} \quad (6)$$

with the function

$$f(s) = \frac{1}{\alpha^2} \frac{x' - 1}{x'} \quad (7)$$

which is related to the axial stress by $N(s) = \mu n^2 \ell^2 f(s) x'(s)$.

An approximate form for $f(s)$ is obtained by linearizing Eq. (3) and neglecting the stretching of the tether. The following expression is obtained

$$f(s) = 3 \left[\frac{\delta}{\beta} + \left(s\delta - \frac{s^2}{2} \right) \right] \quad (8)$$

where $\delta = d/\ell$ is the nondimensional distance of the system center of mass from the SS. As a particular case, for $\beta \rightarrow 0$ is $\delta \rightarrow 0$, $\delta/\beta \rightarrow 1/2 + 1/\gamma$.

Equations (4) show that out-of-plane and in-plane motions are uncoupled and can be solved separately.

In-Plane Vibrations

Solution Procedure

By using matrix notation, Eqs. (4a) and (4b) with corresponding boundary conditions, Eqs. (5a-5d), can be written in the form

$$M\ddot{q} + G\dot{q} + Kq = 0 \quad (9)$$

where $q(s, t)$ is the generalized coordinates vector

$$q(s, t) = \{u(s, t), u(0, t), u(1, t), v(s, t), v(0, t), v(1, t)\}^T \quad (10)$$

The operators M , G , and K are given by

$$\begin{aligned} M &= \begin{bmatrix} m & 0 \\ 0 & m \end{bmatrix}, \quad G = 2 \begin{bmatrix} 0 & -m \\ m & 0 \end{bmatrix} \\ K &= \begin{bmatrix} K_u & 0 \\ 0 & K_v \end{bmatrix} \end{aligned} \quad (11)$$

where

$$m = \text{diag} \left(1, \frac{1}{\beta}, \frac{1}{\gamma} \right) \quad (12a)$$

$$\begin{aligned} K_u &= -\text{diag} \left(3 + \frac{1}{\alpha^2} \frac{d^2}{ds^2}, \frac{3}{\beta} + \frac{1}{\alpha^2} \frac{d}{ds} \right)_{s=0} \\ &\quad \left(\frac{3}{\gamma} - \frac{1}{\alpha^2} \frac{d}{ds} \right)_{s=1} \end{aligned} \quad (12b)$$

$$K_v = -\text{diag} \left(\left[f \frac{d}{ds} \right]', f \frac{d}{ds} \Big|_{s=0}, -f \frac{d}{ds} \Big|_{s=1} \right) \quad (12c)$$

The linear equation [Eq. (9)] admits a solution of the form $q(s, t) = q(s)e^{i\omega t}$, which, after separation of variables, leads to

$$-\omega^2 Mq + i\omega Gq + Kq = 0 \quad (13)$$

Since G is real, $q(s)$ in Eq. (13) must be complex in order to have a nontrivial solution. On account of the particular structure of the matrices involved, it is convenient to partition $q(s)$ into $q = \{u, v\}^T$, where $u = u_r + iu_i$ and $v = v_r + iv_i$. Equation (13) then may be rewritten as four equations,

$$\omega^2 mu_r - 2\omega mv_i - K_u u_r = 0 \quad (14a)$$

$$\omega^2 mv_i - 2\omega mu_r - K_v v_i = 0 \quad (14b)$$

$$\omega^2 mu_i + 2\omega mv_r - K_u u_i = 0 \quad (14c)$$

$$\omega^2 mv_r + 2\omega mu_i - K_v v_r = 0 \quad (14d)$$

where 12 variables are present. It is apparent that Eqs. (14) are coupled two by two and that the two systems of equations are equal if we pose $u_i = u_r$, $v_r = -v_i$. By denoting with z the vector $z = \{u_r, v_i\}^T$, we can rewrite Eq. (13) as

$$\omega^2 Mz - \omega G^* z - Kz = 0 \quad (15)$$

where M and K are the same as before, whereas G^* is now a symmetric matrix defined as

$$G^* = 2 \begin{bmatrix} 0 & m \\ m & 0 \end{bmatrix} \quad (16)$$

After solving Eq. (15), we find that the general solution for the displacement components is given by

$$u(s, t) = \sum_j u_{rj}(s) C_j \sin(\omega_j t + \psi_j) \quad (17a)$$

$$v(s, t) = \sum_j v_{ij}(s) C_j \cos(\omega_j t + \psi_j) \quad (17b)$$

where C_j and ψ_j are arbitrary constants to be determined from initial conditions.

The frequencies of the system can be related to the eigenfunctions u_r and v_i by the following expression

$$\omega^2 = \frac{\int_0^1 v_i^T K_v v_i ds - \int_0^1 u_r^T K_u u_r ds}{\int_0^1 v_i^T m v_i ds - \int_0^1 u_r^T m u_r ds} \quad (18)$$

in which the single terms represent the contribution to potential and kinetic energy of each component of motion. In the following, subscripts r and i will be omitted.

A perturbation technique is now employed to solve the differential equations with variable coefficients [Eq. (15)]. To this end, in the hypothesis $\gamma \leq \mathcal{O}(1)$, $\beta/\gamma \leq \mathcal{O}(1)$, it is possible to consider the function $f(s)$ as the sum of a constant and a variable term

$$f(s) = f_0 + \epsilon f_1(s) \quad (19)$$

with $\epsilon \ll 1$. For example, by using Eq. (8), it is possible to assume $f_0 = f(1) = f_{\min}$ (if $m_2 < m_1$), or $f_0 = f(\delta) = f_{\max}$, or $f_0 = f_{av} = 3(\delta/\beta + \delta/2 - 1/6)$. The influence of the choice of f_0 will be discussed with numerical results.

According to Eq. (19), $K(s)$ can be written as

$$K(s) = K_0 + \epsilon K_1(s) = \begin{bmatrix} K_u & 0 \\ 0 & K_{0v} \end{bmatrix} + \epsilon \begin{bmatrix} 0 & 0 \\ 0 & K_{1v} \end{bmatrix} \quad (20)$$

where

$$K_{0v} = -\text{diag} \left(f_0 \frac{d^2}{ds^2}, f_0 \frac{d}{ds} \Big|_{s=0}, -f_0 \frac{d}{ds} \Big|_{s=1} \right) \quad (21a)$$

$$K_{1v} = -\text{diag} \left(\left[f_1 \frac{d}{ds} \right]', \left[f_1 \frac{d}{ds} \right]_{s=0}, -\left[f_1 \frac{d}{ds} \right]_{s=1} \right) \quad (21b)$$

and K_u is defined by Eq. (12b). Expanding $z(s)$ and ω in terms of ϵ using

$$z(s) = z_0 + \epsilon z_1 + \mathcal{O}(\epsilon^2) \quad (22a)$$

$$\omega = \omega_0 + \epsilon \omega_1 + \mathcal{O}(\epsilon^2) \quad (22b)$$

we obtain the perturbation equations at the order ϵ^0 and ϵ , respectively

$$\omega_0^2 Mz_0 - \omega_0 G^* z_0 - K_0 z_0 = 0 \quad (23)$$

$$\omega_0^2 Mz_1 - \omega_0 G^* z_1 - K_0 z_1 = -2\omega_0 \omega_1 Mz_0 + \omega_1 G^* z_0 + K_1 z_0 \quad (24)$$

At the zeroth order, the equation describes the problem of the TSS with constant stress along the tether. Although the associated eigenvalue problem is not a standard one, particular properties of the eigensolutions can be found. From Eq. (23), written for two different eigenvalues with the corresponding eigenfunctions, performing straightforward manipulations, we get the orthonormalization condition

$$(\omega_{0k} + \omega_{0h}) \int_0^1 z_{0k}^T M z_{0h} ds - \int_0^1 z_{0k}^T G^* z_{0h} ds = \delta_{hk} \quad (25)$$

The first-order solution can be determined from Eq. (24) by writing z_{1j} as a linear combination of the eigenfunctions z_0

$$z_{1j} = \sum_k a_{jk} z_{0k} \quad (26)$$

By applying a standard procedure, we obtain the correction to the frequency ω_{0j} ,

$$\omega_{1j} = \int_0^1 z_{0j}^T K_1 z_{0j} ds \quad (27)$$

where the eigenfunctions z_{0j} have been normalized according to Eq. (25). The coefficients a_{jk} are given by the solution of a linear algebraic coupled system (see Ref. 15).

Explicit Solution of the Eigenvalue Problem

The explicit solution of Eqs. (23) and (24) implies two steps, the first corresponding to the problem with constant stress along the tether (zeroth-order perturbation solution), the second one furnishing the corrections due to the variability of the tension (first-order perturbation solution).

The zeroth-order perturbation, Eq. (23), admits the solution

$$z_0(s) = Ze^{\beta s} \quad (28)$$

By introducing Eq. (28) into Eq. (23), we obtain a fourth-degree characteristic equation whose solutions are

$$\beta^2 = -\frac{1}{2} \left(3\alpha^2 + \omega_0^2 \alpha^2 + \frac{\omega_0^2}{f_0} \right) (1 \pm \sqrt{1 + \zeta}) \quad (29)$$

where

$$\zeta = \frac{4\omega_0^2 \alpha^2 (1 - \omega_0^2)}{f_0 \left(3\alpha^2 + \omega_0^2 \alpha^2 + \frac{\omega_0^2}{f_0} \right)^2} \quad (30)$$

It is possible to show that $(1 + \zeta)$ is positive for any value of ω_0 as it can be written as the sum of two squares. Furthermore, the following inequalities hold $0 > \zeta > -1$ for $\omega_0^2 > 1$, $\zeta = 0$ for $\omega_0^2 = 1$, and $\zeta > 0$ for $1 > \omega_0^2 > 0$; consequently the roots can be real or imaginary. Only the case $\omega_0 > 1$ is analyzed here since $\omega_0 \leq 1$ corresponds to values of the geometric and mechanical parameters, which are out of the ranges of technical interest (see Ref. 15). By defining

$$p^2 = \frac{\alpha^2}{2} \left[3 + \omega_0^2 + \frac{\omega_0^2}{\alpha^2 f_0} \right] (1 + \sqrt{1 + \zeta}) \quad (31a)$$

$$q^2 = \frac{\alpha^2}{2} \left[3 + \omega_0^2 + \frac{\omega_0^2}{\alpha^2 f_0} \right] (1 - \sqrt{1 + \zeta}) \quad (31b)$$

we can write the roots ϑ as $\pm ip$ and $\pm iq$ and the displacement components read

$$u_0(s) = c_p (A_1 \sin ps + A_2 \cos ps) + c_q (A_3 \sin qs + A_4 \cos qs) \quad (32a)$$

$$v_0(s) = A_1 \sin ps + A_2 \cos ps + A_3 \sin qs + A_4 \cos qs \quad (32b)$$

where

$$c_p = \frac{2\omega_0}{3 + \omega_0^2 - p^2/\alpha^2} \quad (33a)$$

$$c_q = \frac{2\omega_0}{3 + \omega_0^2 - q^2/\alpha^2} \quad (33b)$$

By imposing boundary conditions, we get a homogeneous system of linear algebraic equations in the unknown amplitudes A_i ($i = 1, 2, 3, 4$). The vanishing of the determinant of the coefficients leads to a transcendental characteristic equation in the unknown ω_0 , which has to be numerically solved. From this, p and q and, therefore, the solutions $u_0(s)$ and $v_0(s)$ are determined.

The first-order corrections to the frequencies are evaluated easily by means of Eq. (27), which, in this specific case, reads

$$\omega_{1j} = \int_0^1 f_1(s) v_{0j}^2 ds \quad (34)$$

where $v_{0j}(s)$ is furnished by Eq. (32b).

The eigenfunctions' corrections are determined by numerically solving the system for coefficients a_{jk} (see Ref. 15) where equations are coupled, since eigenfunctions are not orthogonal in the classical sense defined for natural systems, i.e., systems with no gyroscopic forces.

Out-of-Plane Vibrations

The out-of-plane motion has been studied in Ref. 12; the relevant results are summarized here. The problem is governed by Eq. (4c) with the boundary conditions of Eqs. (5e) and (5f). By posing $w(s, t) = w(s)e^{i\omega t}$, we can write them in the matrix form

$$(1 - \Omega^2)Mw + Kw = 0 \quad (35)$$

where $M = m$, $K = K_v$, as defined previously for the in-plane motion, and $w = \{w(s), w(0), w(1)\}^T$. The solution of the problem is obtained through the perturbation procedure previously employed by expanding Ω^2 and w into series of the small parameter ϵ appearing in Eq. (19). The relevant perturbation equations are

$$(\Omega_0^2 - 1)Mw_0 - K_0w_0 = 0 \quad (36)$$

$$(\Omega_0^2 - 1)Mw_1 - K_0w_1 = -\Omega_1^2 Mw_0 + K_1w_0 \quad (37)$$

where K_0 and K_1 stand for K_{0v} and K_{1v} , defined by Eqs. (21a) and (21b).

The solution of Eq. (36) is given by

$$w_0(s) = A_1 \sin \lambda s + A_2 \cos \lambda s \quad (38)$$

where $\lambda^2 = (\Omega_0^2 - 1)/f_0$ has been posed. By using boundary conditions, we obtain the following characteristic equation:

$$\tan \lambda = \frac{\beta + \gamma}{\lambda^2 - \beta\gamma} \lambda \quad (39)$$

The eigenfunctions w_{0j} verify the classical orthonormality conditions for natural systems and are independent of the choice of the constant stress value.

The first-order perturbation, Eq. (37), is solved by assuming $w_{1j} = \sum_k a_{jk} w_{0k}$. The frequency correction obtained is

$$\Omega_{1j}^2 = \int_0^1 f_1(s) w_{0j}^2 ds \quad (40)$$

from which the natural frequencies, up to the first order, are

$$\Omega_j^2 = \Omega_{0j}^2 + \epsilon \Omega_{1j}^2 = 1 + \int_0^1 f(s) w_{0j}^2 ds \quad (41)$$

where use has been made of Eq. (19) and of the orthonormalization conditions. The coefficients a_{jk} are furnished by

$$a_{jk} = \frac{1}{\Omega_j^2 - \Omega_k^2} \int_0^1 f(s) w_{0k}' w_{0j}' ds \quad (42)$$

It should be noticed that, whereas eigenfunctions w_j depend on the choice of f_0 , Ω_j^2 is independent of the constant value of the stress, which only affects the frequency approximation at order ϵ^0 .

Asymptotic Analysis of the In-Plane Motion

As already observed, the two components of the in-plane motion, $u(s, t)$ and $v(s, t)$, are coupled by means of the gyroscopic forces; a measure of the coupling is given by the coefficients c_p and c_q appearing in Eqs. (32). In particular, the solution depends on the nondimensional parameter α^2 , defined by Eqs. (6), which, in the range of technical values of the geometric and mechanical properties of the system, assumes very small values, at the most of order $\mathcal{O}(10^{-4})$. Therefore, it is worth examining the order of magnitude of the different terms of the solution in order to point out the coupling between the components of motion and to obtain an approximate solution.

To this end, two different approaches to the problem are presented. In the first one, the asymptotic behavior of the solution for small values of α^2 is studied; in the second one, the general equations of motion [Eqs. (4a) and (4b)] are solved by means of a perturbation technique in which α^2 is assumed as the perturbation parameter. In both cases, for simplicity, we refer to an infinite mass of the SS ($\beta = 0$). Note that the coefficients γ and $f(s)$ are both of order $\mathcal{O}(1)$.

Asymptotic Expansion of the Constant Stress Solution

The eigenvalue problem of in-plane motion, related to the case $\beta = 0$, can be written more explicitly as

$$\begin{bmatrix} a_{11} & a_{12} \\ a_{21} & a_{22} \end{bmatrix} \begin{Bmatrix} A_1 \\ A_3 \end{Bmatrix} = \begin{Bmatrix} 0 \\ 0 \end{Bmatrix} \quad (43)$$

where

$$a_{11} = \frac{c_p p}{2\alpha^2 \omega_0} (p \sin p - \gamma \cos p) \quad (44a)$$

$$a_{12} = \frac{c_q q}{2\alpha^2 \omega_0} (q \sin q - \gamma \cos q) \quad (44b)$$

$$a_{21} = \left(1 - \frac{2c_p}{\omega_0}\right) \omega_0^2 \sin p - \gamma f_0 p \cos p \quad (44c)$$

$$a_{22} = \left(1 - \frac{2c_q}{\omega_0}\right) \omega_0^2 \sin q - \gamma f_0 q \cos q \quad (44d)$$

being $A_2 = A_4 = 0$ from the boundary conditions $u(0) = v(0) = 0$. In Eqs. (44), the coefficients c_p and c_q and the spatial frequencies p and q are functions of α^2 according to Eqs. (30), (31), and (33). By expanding $\sqrt{1+\xi}$ in terms of α^2 and substituting into Eqs. (31), we get, up to the second order

$$p^2 = \frac{\omega_0^2}{f_0} + 4\alpha^2 \quad (45a)$$

$$q^2 = -\alpha^2 \left(1 - \omega_0^2\right) \left(1 + 4\alpha^2 \frac{f_0}{\omega_0^2}\right) \quad (45b)$$

from which, by introducing into Eqs. (33),

$$c_p = -\frac{2\alpha^2 \omega_0^2}{p^2} \quad (46a)$$

$$c_q = \frac{\omega_0}{2} \left[1 + \alpha^2 \frac{f_0}{\omega_0^2} \left(1 - \omega_0^2\right)\right] \quad (46b)$$

In view of these results, it is found that the eigenvalue problem [Eq. (43)] admits the ∞^2 solutions

$$A_1 \neq 0, \quad A_3 = 0, \quad \tan p = \gamma/p \quad (47a)$$

$$A_1 = 0, \quad A_3 \neq 0, \quad \tan q = \gamma/q \quad (47b)$$

Equations (47) show that the spatial frequencies p and q are equal in that they are the solution of the same characteristic equation and, furthermore, they coincide with the eigenvalues λ of the out-of-plane motion. Therefore, it emerges from Eqs. (45a) and (38) that out-of-plane and in-plane frequencies up to order α verify the relation $\omega_0^2 = \Omega_0^2 - 1$. Moreover, from Eqs. (47), it appears that the harmonics in p and q are uncoupled. By accounting for Eqs. (46) and (47), we find that the vibration modes [Eqs. (32)] are

$$u_0(s) = -\frac{2\alpha^2 \omega_0^2}{p^2} A_1 \sin ps \quad (48a)$$

$$v_0(s) = A_1 \sin ps \quad (48b)$$

or

$$u_0(s) = \frac{\omega_0}{2} A_3 \sin qs \quad (49a)$$

$$v_0(s) = A_3 \sin qs \quad (49b)$$

In both cases, it ensues that $u_0(s)/v_0(s) = \text{const.}$

Equations (48) show that because of the presence of α^2 in Eq. (48a) the two components of motion are weakly coupled and, besides, the transverse component prevails over the axial component. For this reason, we shall refer to Eqs. (48) as *prevalently transverse* eigenfunctions. Because of Eq. (45a), the first vibration modes associated with values of p of order $\Theta(1)$ correspond to frequencies ω_0 of the same order.

From Eqs. (49), it ensues that $u_0(s)/v_0(s) = \omega_0/2$ and, therefore, the longitudinal component of the motion is dominant with respect to the transverse one since frequencies are of order $\Theta(\alpha^{-1})$. This is seen from Eq. (45b) if one considers that the first q is of order $\Theta(1)$. Consequently, these modal shapes will be referred to as *prevalently longitudinal* eigenfunctions, which are characterized by higher values of the frequencies in comparison with the prevalently transverse ones associated with the same values of p .

In conclusion, in both cases the coupling between the two components of motion due to the Coriolis forces is rather weak.

Asymptotic Solution of the General Equations of Motion

Let us consider the linearized equations of the in-plane motion [Eqs. (4a) and (4b)] and the corresponding boundary conditions [Eqs. (5a-5d)] where $\beta = 0$.

An asymptotic solution for $\alpha^2 \rightarrow 0$ is sought. To this end, attention is concentrated on modal shape wavelengths of order $\Theta(1)$ for which $u'' = \Theta(u)$ and $v'' = \Theta(v)$ and a distinction is made between two cases according to the value of ω :

1) $\omega = \Theta(1)$: In this case, frequencies are of the same order as those of the transverse oscillations of a string. $\ddot{u} = \Theta(u)$, $\ddot{v} = \Theta(v)$, and similarly for v .

2) $\omega = \Theta(\alpha^{-1})$: In this case, frequencies are of the same order as those of the longitudinal oscillations of a string. $\ddot{u} = \Theta(\alpha^{-2}u)$, $\dot{u} = \Theta(\alpha^{-1}u)$, and similarly for v .

The two cases are treated separately.

Transverse Type Modes

By performing a series expansion in terms of the perturbation parameter α^2 , we may write the solution of Eqs. (4a) and (4b) as

$$u = u_0 + \alpha^2 u_1 \quad (50a)$$

$$v = v_0 + \alpha^2 v_1 \quad (50b)$$

from which the perturbation equations with relevant boundary conditions are derived:

Order α^0 :

$$u_0'' = 0 \quad (51)$$

$$u_0(0, t) = 0 \quad (52a)$$

$$u_0'(1, t) = 0 \quad (52b)$$

Order α^2 :

$$-2\dot{v}_0 - u_1'' = 0 \quad (53a)$$

$$\ddot{v}_0 - [f(s)v_0']' = 0 \quad (53b)$$

$$u_1(0, t) = 0 \quad (54a)$$

$$v_0(0, t) = 0 \quad (54b)$$

$$-2\dot{v}_0(1, t) + \gamma u_1'(1, t) = 0 \quad (54c)$$

$$\ddot{v}_0(1, t) + \gamma f(1)v_0'(1, t) = 0 \quad (54d)$$

The solution of Eqs. (51) and (52) furnishes $u_0(s, t) = 0$ and, therefore, $u = \Theta(\alpha^2)$. Equation (53b) describes the vibrations of a string under variable tension $f(s)$ and admits a solution of the type

$$v_0(s, t) = v_0(s) \cos(\omega t + \psi) \quad (55)$$

where $v_0(s)$ and ω have to be determined by imposing the boundary conditions [Eqs. (54b) and (54d)]. If $f(s)$ is approx-

imated with a constant value f_0 , for instance, its average value, the solution is given by $v_0(s) = A \sin ps$. The associated value of the frequencies is

$$\omega_0 = \sqrt{f_0} p \quad (56)$$

where p is the solution of the characteristic equation

$$\tan p = \gamma/p$$

By performing integrations on Eq. (53a) and by using the characteristic equation and appropriate boundary conditions, we obtain $u(s, t)$. Therefore, at the present level of the perturbation procedure, the solution reads

$$u(s, t) = -2\alpha^2 \frac{\omega_0}{p^2} A \sin ps \sin(\omega_0 t + \psi) \quad (57a)$$

$$v(s, t) = A \sin ps \cos(\omega_0 t + \psi) \quad (57b)$$

Within this approximation, the following results have been achieved.

1) The frequency coincides with that of the transverse oscillations of a string and it is not modified by the longitudinal motion.

2) The amplitude of the longitudinal displacement $u(s, t)$ is small of order $O(\alpha^2)$ in comparison with the transverse one.

3) The displacement components $u(s, t)$ and $v(s, t)$ oscillate with a phase difference of $\pi/2$.

Note that the frequencies and the modal shapes coincide with those given by the series expansion of the exact solution of the coupled oscillations with constant stress.

The employed perturbation procedure is susceptible to a useful interpretation that helps understanding of the mechanical behavior of the system. The equation that governs the most important aspects of the phenomenon is Eq. (53b), which, as previously stated, corresponds to a purely transverse motion. Equation (53a) represents the equilibrium condition in the longitudinal direction of the string where u'' and \dot{v}_0 are, respectively, proportional to the elastic reaction and to the longitudinal component of the Coriolis force generated by the transverse motion. A stress increment, which is proportional to the linearized stretching u' , arises in the string to balance, on the section of abscissa s , the Coriolis' forces acting on the cable from the current section up to the free end as well as the Coriolis' force on the satellite. It can be seen that the stress increment is small in comparison with the prestress.

At the next order, which is not examined here, the longitudinal displacements appear in the transverse equilibrium equation by producing a Coriolis' force proportional to \dot{u} , responsible for a small frequency correction. In conclusion, the procedure shows that in the prevalently transverse motion the axial displacement of the cable represents only a secondary aspect of the phenomenon if the cable is characterized by sufficiently high axial rigidity.

Longitudinal Type Modes

In this case, it is convenient to modify the time scale by introducing a new variable $\tau = \alpha^{-1}t$. It is, therefore, $\omega t = \omega \alpha \tau = \bar{\omega} \tau$ where $\bar{\omega} = \omega \alpha = O(1)$. In the new time scale, it is $\partial u(s, \tau)/\partial \tau = O(\bar{\omega} u) = O(u)$, $\partial^2 u(s, \tau)/\partial \tau^2 = O(\bar{\omega}^2 u) = O(u)$.

By expanding u and v in series of α , i.e.,

$$u = u_0 + \alpha u_1 \quad (58a)$$

$$v = v_0 + \alpha v_1 \quad (58b)$$

we obtain the relevant perturbation equations through standard steps. By solving the equations and coming back to variable t , we obtain

$$u(s, t) = A \sin ps \sin(\omega t + \psi) \quad (59a)$$

$$v(s, t) = (2/\omega) A \sin ps \cos(\omega t + \psi) \quad (59b)$$

where $\omega^2 = q^2/\alpha^2$ and q satisfies the characteristic equation $\tan q = \gamma/q$. It is interesting to observe that the characteristic equation coincides with Eqs. (47b), Eqs. (59) are in accordance with Eqs. (49), and, besides, ω^2 is a good approximation of Eq. (45b) since $\omega \gg 1$. Hence, in this case too, the solution previously obtained is recovered.

Numerical Results

The main results presented here regard the analysis of a TSS with $m_1 \gg m_2$; some results relative to systems with equal concentrated masses will be illustrated at the end of the section. Let us assume the orbital angular velocity $n^2 = 1.35 \cdot 10^{-6} \text{ s}^{-2}$, orbital radius $a = 6657 \text{ km}$, SS mass $m_1 = 10^5 \text{ kg}$, satellite mass $m_2 = 500 \text{ kg}$, tether mass $\mu = 5.76 \cdot 10^{-3} \text{ kg/m}$, and tether cross section $A = 4 \cdot 10^{-6} \text{ m}^2$, Young's modulus $E = 7 \cdot 10^{10} \text{ N/m}^2$, and tether length ℓ is a variable. Two systems are analyzed in detail corresponding to $\ell = 20$ and 100 km , for which nondimensional parameters assume the values $\alpha^2 = 1.111 \cdot 10^{-5}$, $\beta = 0.001152$, $\gamma = 0.2304$, $\delta = 0.0055$, and $\alpha^2 = 2.777 \cdot 10^{-4}$, $\beta = 0.00576$, $\gamma = 1.152$, $\delta = 0.0078$, respectively. The analysis has been performed by referring to the solution reported in the third section and results are compared with those of the asymptotic solution.

Table 1 collects the in-plane frequencies for $\ell = 100 \text{ km}$ corresponding to three values of f_0 , namely, the lowest ($f_{\min} = 2.585$), the average ($f_{\text{av}} = 3.574$), and the largest ($f_{\max} = 4.062$) value of $f(s)$. It is noticed that, whereas ω_0 usually grows with f_0 , the values of $\omega = \omega_0 + \epsilon \omega_1$ are practically independent of f_0 . Besides, the corrections ω_1 relative to the average value of $f(s)$ are very small and decrease with the mode number, thus giving $\omega \approx \omega_0$. Similar results are obtained for a length of 20 km .¹⁵

A hypothesis that permits simplification of the numerical analysis consists in assuming that the SS mass is infinite, which implies $\beta = 0$. Under this assumption, apart from the first modes, results change very little. For decreasing lengths, the difference between results corresponding to $\beta = 0$ and $\beta \neq 0$ tends to disappear.

Numerical results, confirmed by the asymptotic analysis, show that the vibration modes are characterized by a weak coupling between the two components of motion $u_0(s)$ and $v_0(s)$, and, therefore, they are of prevalently transversal or longitudinal type. In particular, among the first modes, most are of transversal type except for a few, for instance, $i = 11, 37, \dots$ for $\ell = 100 \text{ km}$ or $i = 13, \dots$ for $\ell = 20 \text{ km}$. The corresponding frequencies ω_0 grow with f_0 in the prevalently transversal modes since they are related to the geometric stiffness or they remain practically constant in the prevalently longitudinal ones because of their dependence on EA (see Table 1). It is found that harmonics with circular frequency p dominate over those in q in the prevalently transversal modes, whereas the opposite occurs in the prevalently longitudinal modes. This has been pointed out by the asymptotic analysis according to which transverse modes contain only harmonics in p [Eqs. (48) and (57)] and longitudinal modes contain only

Table 1 Time frequencies of in-plane oscillations ($\ell = 100 \text{ km}$)

Mode	f_{\min}		f_{av}		f_{\max}	
	ω_0	ω	ω_0	ω	ω_0	ω
1	1.467	1.819	1.725	1.794	1.840	1.797
2	5.590	7.046	6.577	6.905	7.018	6.900
3	10.426	12.672	12.265	12.475	13.088	12.488
4	15.404	18.534	18.122	18.269	19.338	18.299
5	20.426	24.480	24.030	24.143	25.643	24.187
11	54.559	54.559	54.559	54.555	54.559	54.559
37	207.883	207.883	207.883	207.883	207.883	207.883

those in q [Eqs. (49) and (59)]. From this, it follows that the motion components $u_0(s)$ and $v_0(s)$ can be assumed proportional, with constant ratio equal to c_p and c_q , respectively.

Figures 2 show plots of the transversal and longitudinal displacements relative to mode numbers 1–4, 11, and 37 for $\ell=100$ km and mode numbers 1–5 and 13 for $\ell=20$ km, evaluated for $f_0=f_{av}$ in both cases. The values of the ratio u_0/v_0 reported in Figs. 2 confirm that the motion components are practically uncoupled. The first modes in Figs. 2a and 2b are of a pendular type, whereas the subsequent modes are similar to those of a vibrating string, except for the presence of a small displacement at the end $s=1$ where the satellite is attached; also at $s=0$ a small displacement of the SS occurs, especially for great lengths of the tether. The behavior of $u_0(s)$ at prevalently longitudinal modes is similar to that of $v_0(s)$ at corresponding prevalently transversal ones, although ω_0 is much higher.

The values of p and q are collected in Table 2 for $\ell=100$ km. Each frequency p of the prevalently transversal modes is matched by one frequency q of the prevalently longitudinal modes, and, therefore, the wavelengths of the two modes are approximately multiples of π , except for the first modes. It is found that these values are independent of the choice of the constant value f_0 , according to Eqs. (47). Besides, they are practically the same as frequencies λ of out-of-plane modes, also listed in Table 2.

Previous results regarding the weak coupling between $u_0(s)$ and $v_0(s)$ are confirmed by evaluating the kinetic and potential

energy terms present in Eq. (24). Table 3 shows that the energy terms relative to $u_0(s)$ are small in comparison with those relative to $v_0(s)$ in the prevalently transversal modes, whereas they become dominant in the prevalently longitudinal modes.

Figure 3 shows a plot of the time frequencies vs the mode number for different lengths of the tether. It is apparent that, for each value of the length, the behavior is practically linear, with the exception of the first modes; besides, the time frequencies tend to be independent of the length as it increases. This is confirmed by considering the perturbation equation [Eq. (53b)] (no gyroscopic forces) and Eq. (8). It is seen that for $\ell \rightarrow \infty$, $f(s) \rightarrow 3(s-s^2)/2$, i.e., the tension is length independent; consequently, time frequencies tend to be a limit value different from zero.

By enlarging the abscissa scale, we notice that the original straight line associated with each value of the length translates to parallel paths by correspondence with mode numbers $i=11, 37, \dots$ for $\ell=100$ km or $i=13, \dots$ for $\ell=20$ km; these jumps are associated with prevalently longitudinal modes. Figure 4 and Table 2 illustrate this situation together with the behavior of out-of-plane time frequencies Ω (straight lines in Fig. 4). Note that the amount of the translation, measured along the horizontal line, is equal to one mode and the same difference occurs between in-plane and out-of-plane spatial frequencies.

The coefficients a_{jk} relative to the shape corrections of the eigenfunctions of in-plane vibrations have been determined by solving the appropriate system of equations. It is found that these equations are practically uncoupled since the ratio between out-of-diagonal and diagonal coefficients is at the most of order $\mathcal{O}(10^{-3})$, because the wavelengths of the eigenfunctions are approximately multiples of π . Table 4 shows the values of a_{jk} calculated by considering for f_0 , the average value of $f(s)$. It is apparent that these values rapidly decrease starting from the mode to be corrected so that, in the analysis, it is sufficient to consider a narrow band around the mode in question. Similar results are obtained for out-of-plane motion (see Ref. 15).

Results obtained so far are compared with those illustrated in Ref. 10. The present solution depends on three nondimen-

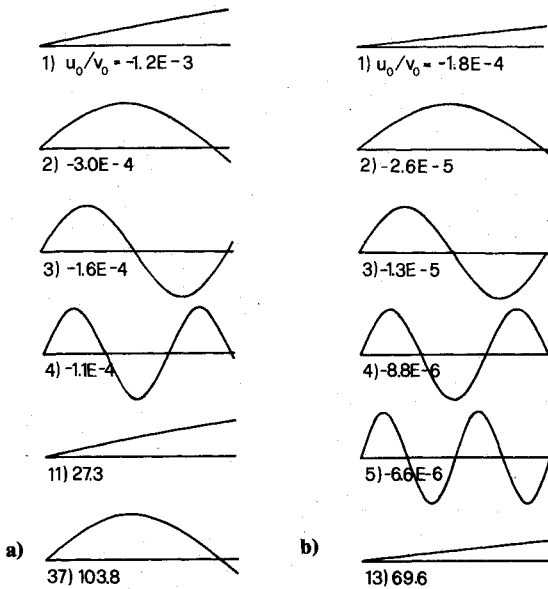


Fig. 2 In-plane modal shapes: a) $\ell=100$ km; b) $\ell=20$ km.

Table 2 Time and spatial frequencies of in-plane and out-of-plane oscillations ($f_0=f_{av}$, $\ell=100$ km)

Mode	In-plane oscillations			Out-of-plane oscillations	
	ω	p/π	q/π	Ω	λ/π
1	1.794	0.289	0.007	2.006	0.289
2	6.905	1.103	0.034	6.587	1.103
3	12.457	2.056	0.065	12.263	2.056
4	18.269	3.038	0.096	18.119	3.038
5	24.143	4.029	0.127	24.027	4.029
10	53.807	9.013	0.285	53.754	9.013
11	54.559	9.148	0.289	59.724	10.012
12	59.758	10.012	0.317	65.680	10.823
37	207.883	34.855	1.103	214.734	36.003

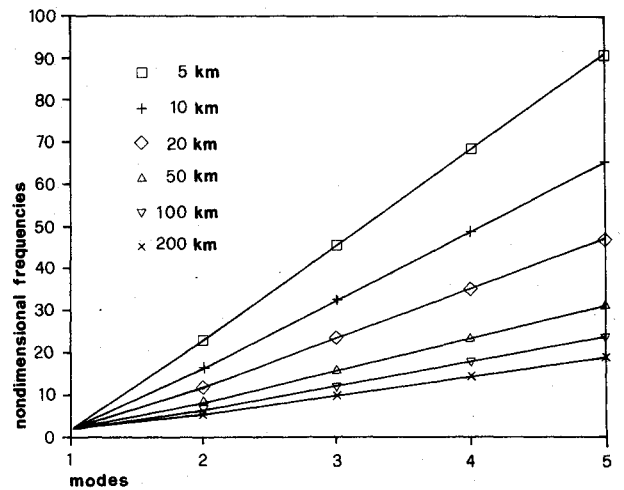


Fig. 3 In-plane time frequencies for different lengths of the tether.

Table 3 Energy contributions of u_0 and v_0 ($\ell=100$ km)

Mode	Kinetic u_0	Potential u_0	Kinetic v_0	Potential v_0
1	3.91E-7	1.16E-3	2.90E-1	8.62E-1
2	7.06E-9	3.05E-4	7.60E-2	3.29E+0
3	1.09E-9	1.64E-4	4.08E-2	6.13E+0
4	3.37E-10	1.11E-4	2.76E-2	9.06E+0
5	1.46E-10	8.35E-5	2.08E-2	1.20E+1
11	9.16E-3	2.72E+1	1.23E-5	3.68E-5
37	2.41E-3	1.04E+2	2.23E-7	9.65E-6

Table 4 Coefficients a_{jk} of the shape corrections of in-plane modes ($\ell=100$ km)

Mode	$j-3$	$j-2$	$j-1$	$j+1$	$j+2$	$j+3$
1	—	—	—	$-6.56\text{E}-2$	$9.44\text{E}-3$	$-3.21\text{E}-3$
2	—	—	$1.06\text{E}-2$	$-9.44\text{E}-2$	$1.83\text{E}-2$	$-6.92\text{E}-3$
3	—	$-8.81\text{E}-4$	$4.66\text{E}-2$	$-1.36\text{E}-1$	$2.30\text{E}-2$	$-8.55\text{E}-3$
4	$2.86\text{E}-4$	$-5.26\text{E}-3$	$9.06\text{E}-2$	$-1.76\text{E}-1$	$2.78\text{E}-2$	$-1.00\text{E}-2$
5	$1.47\text{E}-3$	$-1.13\text{E}-2$	$1.32\text{E}-1$	$-2.17\text{E}-1$	$3.28\text{E}-2$	$-1.15\text{E}-2$

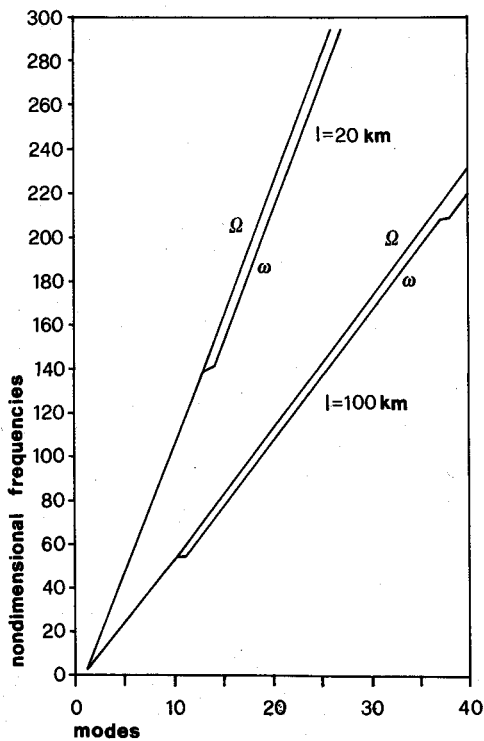


Fig. 4 Time frequencies of in-plane and out-of-plane oscillations.

Table 5 In-plane and out-of-plane frequencies ($\alpha^2=1.111 \cdot 10^{-5}$, $\beta=0$, $\gamma=0.2$)

Mode	Present theory				Reference 10	
	ω_0	ω	Ω_0	Ω	ω	Ω
1	1.731	1.733	1.999	2.000	1.732	2.000
2	12.816	12.780	12.854	12.819	12.776	12.815
3	25.259	25.241	25.279	25.261	25.174	25.193
4	37.784	37.771	37.797	37.784	37.760	37.773
5	50.329	50.319	50.339	50.329	50.305	50.315

Table 6 In-plane and out-of-plane frequencies ($\alpha^2=1.111 \cdot 10^{-5}$, $\beta=0$, $\gamma=1$)

Mode	Present theory				Reference 10	
	ω_0	ω	Ω_0	Ω	ω	Ω
1	1.721	1.742	1.990	2.000	1.732	2.000
2	6.851	6.750	6.294	6.823	6.707	6.782
3	12.875	12.811	12.913	12.850	12.742	12.782
4	19.059	19.014	19.085	19.040	18.915	18.942
5	25.291	25.256	25.310	25.276	25.127	25.147

Table 7 In-plane frequencies and displacements ($\ell=20$ km)

Mode	$\beta=0.001152, \gamma=0.23$			$\beta=\gamma=0.001152$			$\beta=\gamma=0.2304$		
	ω	$ v_1/v_{\max} $	$ v_2/v_{\max} $	ω	$ v_{1,2}/v_{\max} $	ω	$ v_{1,2}/v_{\max} $	ω	$ v_{1,2}/v_{\max} $
1	1.738	$5.58\text{E}-3$	1.0	1.728	1.0	1.742	1.0		
2	11.993	$3.58\text{E}-4$	$7.15\text{E}-2$	113.385	$3.66\text{E}-4$	8.341	$7.00\text{E}-2$		
3	23.644	$1.82\text{E}-4$	$3.64\text{E}-2$	226.736	$1.83\text{E}-4$	16.191	$3.62\text{E}-2$		
4	35.368	$1.21\text{E}-4$	$2.43\text{E}-2$	340.095	$1.22\text{E}-4$	24.155	$2.43\text{E}-2$		
5	47.112	$0.92\text{E}-4$	$1.83\text{E}-2$	453.455	$0.92\text{E}-4$	32.142	$1.82\text{E}-2$		

sional parameters α^2, β, γ [Eqs. (6)], whereas the solution in Ref. 10 depends on the unique parameter γ . This is because, in were neglected, thus implying the transversal modes to be independent of α^2 . On the other hand, the solution in Ref. 10 is more accurate than the present one since Legendre polynomials are employed. Tables 5 and 6 collect the first five in-plane ω and out-of-plane Ω frequencies of two systems with $\alpha^2=1.111 \cdot 10^{-5}$, $\beta=0$, and $\gamma=0.2$ and 1, respectively. When $\gamma=0.2$, it is seen that even zeroth-order frequencies ω_0 and Ω_0 are in good accord with result of Ref. 10; the agreement is far better for first-order frequencies. When $\gamma=1$, the approximation is not as good, but still acceptable. For increasing γ , differences increase because the stress $f(s)$ is no more slowly variable, and, therefore, the hypothesis of applicability of the perturbation method fails.

Finally, the influence of the lumped masses m_1 and m_2 on the frequencies and modal shapes of the system are analyzed. Three TSS are considered and results relative to the in-plane

transversal modes are compared. The first system is one of those previously considered, with $m_1=10^5$ kg, $m_2=500$ kg, whose modal shapes are plotted in Fig. 2b; in the second system, $m_1=m_2=10^5$ kg, in the third one, $m_1=m_2=500$ kg. The mass of the tether is the same in the three cases. The first-order frequencies ω displayed in Table 7 are seen to be strongly dependent on the increase of the masses because of the increasing of the stress, except for the pendular mode whose frequency is stress independent. The ratios between the two end displacements v_1 and v_2 and the maximum displacement v_{\max} are also shown in the table. It is seen that, starting from the second mode, when the masses are both large, the modal shapes are nearly sinusoidal with zero displacement at the ends; when the masses decrease, the displacement becomes noticeable. It is interesting to note that the displacement at one end is nearly independent of the value of the mass at the other end, especially for the higher modes. Finally, the first mode of the TSS with equal masses is an approximation of the exact

solution consisting of a rigid rotation about the center of mass, located at midspan.

Conclusions

The free oscillations of the tethered satellite system have been investigated by utilizing a continuum elastic model. The general solution of the linearized equations of motion has been determined through an analytic procedure by first solving a constant stress problem and then perturbing its solution. Numerical results have shown that a good approximation of the solution is achieved by considering, at the first step, the average value of the tension. An asymptotic solution has also been obtained for the in-plane motion that is in accordance with the previous solution for technical values of the parameters of the system.

Results have shown that the two in-plane motion components are weakly coupled through the gyroscopic forces and the modes can be distinguished in prevalently transversal or prevalently longitudinal ones, with approximately the same wavelength. Besides, the out-of-plane modes are characterized by wavelengths quite equal to that of the prevalently transversal modes and time frequencies Ω related to in-plane ones ω by $\Omega^2 = \omega^2 + 1$.

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